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VOLUME II

NSWC

TECHNICAL REPORT

WHITE OAK LABORATORY

HANDBOOK OF INVISCID SPHERE-CONE FLOW FIELDS AND PRESSURE DISTRIBUTIONS VOLUME II

1 DECEMBER 1975

NAVAL SURFACE WEAPONS CENTER
WHITE OAK LABORATORY
SILVER SPRING, MARYLAND 20910

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This report contains numerical tables of aerodynamic coefficients and surface pressure distributions for sphere cone configurations as a function of angle of attack and Mach number. These results were generated by an NSWC/WOL computer code based on a finite difference solution of the steady inviscid three-dimensional compressible flow equation for a perfect gas $\gamma = 1.4$. Cone half angles of 5°, 6°, 7°, 8°, 9°, 10° 15° and 20° are considered. Truncated values are obtained ~~at axial stations~~ to

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a length of 200 times the sphere radius, over a Mach number range of 3.5, 5, 10, 15, 20, 25 and 30, for an angle-of-attack span of 1^{o}, 3^{o}, 5^{o}, and 10^{o}. Center-of-pressure location, and axial- and normal-force coefficients are tabularly presented at each axial-length increment station. Surface pressure distributions are also presented at each axial-length increment station, including seven meridian plane angles over the half plane from windward to leeward ray. The techniques utilized in generating the tables are described, and comparisons between computed values and values measured in some wind-tunnel experiments are presented. The presentation format is discussed to establish the mechanics necessary to use the tables. The tables are divided into two volumes; the first containing pressure information, and the second, aerodynamic information.

NSWC/WOL/TR 75-45 1 December 1975

HANDBOOK OF INVISCID SPHERE-CONE FLOW FIELDS AND PRESSURE DISTRIBUTIONS - VOLUME II

This report contains numerical tables of surface-pressure distribution and aerodynamic coefficients for sphere cone configurations over a range of Mach numbers and angles of attack relevant to typical re-entry environments. These results were generated by an NSWC/WOL computer code based on a finite difference solution, of the steady inviscid three-dimensional compressible flow equations for a perfect gas $\gamma = 1.4$. This work was sponsored by Naval Sea Systems Command and carried out under the Aeroballistic Re-Entry Technology Program, L. Pasiuk, NAVSEA 035 Manager. This project was performed under task number SF 3232250F.

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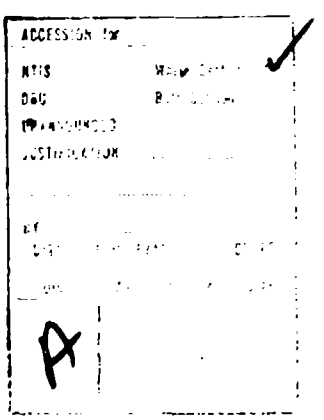
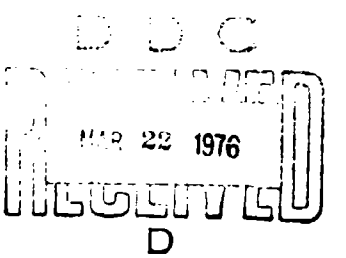
Kurt R. Enkenhus

KURT R. ENKENHUS
By direction





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VOLUME II: AERODYNAMICS

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Symbols - Volume II

A	axial force
b	body radius
b_Z	body slope, $\frac{db}{dz}$
CA	axial force coefficient, A/qR_B^2
CN	normal force coefficient, N/qR_B^2
D	sphere cone base diameter, $2R_B$
L	length measured from sphere-cone nosetip
LV	length measured from virtual sharp cone nosetip
M_∞	freestream Mach No.
N	normal force
P	surface pressure
p_∞, $P_{FREESTREAM}$	freestream pressure
q	freestream dynamic pressure, $\frac{1}{2}\rho_\infty V_\infty^2 = \frac{1}{2}\frac{p_\infty}{\gamma}M_\infty^2$
RB	sphere cone base radius
RN	sphere cone nose radius
XCP	center-of-pressure location measured from sphere cone nosetip
XVCP	center-of-pressure location measured from virtual sharp cone nosetip
YCP	center-of-pressure location measured from sphere cone base
θ_C	sphere cone half angle
ρ_∞	freestream density

ILLUSTRATIONS

INTRODUCTION

The design of re-entry configurations requires a source for the aerodynamic performance characteristics used in predicting the ballistic trajectory. It has proven difficult to obtain such data on sphere cones at angle of attack in the open literature. Some data are available as the result of experimental investigations, and other data are available as the result of study contracts employing the method of characteristic or finite difference techniques to determine the pressure field on particular shapes. Early work by the Russians[1] and subsequent modifications[2] arranged in tabular form have proven most useful for trajectory calculation, heat-transfer calculations, etc.

This report gives, in tabular form, the inviscid surface pressure distributions and the aerodynamic characteristics for sphere-cones at angle of attack. The ranges of the tables are

$$0 \leq \alpha \leq 10^{\circ}$$
$$5^{\circ} \leq \theta_c \leq 20^{\circ}$$
$$3.5 \leq M_{\infty} \leq 30$$
$$\text{perfect gas, } \gamma = 1.4$$

The term cone angle used herein refers to the angle between the surface of the cone and its axis of symmetry. This value is sometimes called the half-cone angle or semi-cone angle. The report is divided into two volumes; Volume I contains the tabulated surface pressure data, Volume II contains the aerodynamic data.

All the tabulated data presented in this report were obtained using an inviscid flow computer code developed at NAVSURFWPNCEN/WOL.[3] A brief description of this code is included and some comparisons are made with experimental results (see Volume I). The arrangement of the tables (both Volumes) is described in the section "Using the Tables."

---

[1] Tables of Supersonic Flow About Blunted Cones, Academy of Sciences, USSR (Moscow), prepared by P. I. Chushkin and W. P. Shuhshmina from: Computation Center Monograph, 1961; translated and edited by J. F. Springfield, Research and Advanced Development Division, AVCO Corp., Wilmington, Mass., RAD-TM-62-63, Sep 1962

[2] Pressure Distributions on Sphere Cones, D. M. Ellett, SC-RR-64-1796, Sandia Laboratory, Albuquerque, N. M., Jan 1965

[3] Solomon, J. M., Ciment, M., and Ferguson, R. E., documentation in progress

INTEGRATION OF PRESSURE DISTRIBUTIONS

Aerodynamic Coefficients

The aerodynamic coefficients presented herein are defined by:

$$CA = 2\int_0^L b_z b \left\{ \int_0^{\pi} p \; d\phi \right\} dz/\pi q \; R_B^2$$

$$CN = 2\int_0^L b \left\{ \int_0^{\pi} p \cos \phi \; d\phi \right\} dz/\pi q R_B^2$$

$$XCP = \int_0^L b(z+b_z b) \left\{ \int_0^{\pi} p \cos \phi \right\} dz \Big/ \int_0^L b \left\{ \int_0^{\pi} p \cos \phi \; d\phi \right\} dz$$

where CA and CN are the axial- and normal-force coefficients, respectively, and XCP is the center of pressure measured from the body nose. For the definitions of the other quantities appearing in the above see the symbols list and Figure 7.

The integrals appearing in these definitions were evaluated numerically using the computed surface pressure distributions tabulated in Volume I. The integrals with respect to ϕ were evaluated using Simpson's rule and the integrals with respect to z were evaluated using the trapezoidal rule.

USING THE TABLES: AERODYNAMIC COEFFICIENTS

The aerodynamic data are divided according to angle of attack and then are subdivided according to cone angle and then according to Mach number. Angles of attack of 1^o, 3^o 5^o, and 10^o are considered over a Mach number range of 3.5, 5.0, 10.0, 15.0, 20.0, 25.0, and 30.0. Cone angles of 5^o, 6^o, 7^o, 8^o, 9^o, 10^o, 15^o and 20^o are utilized. Each table is headed by the appropriate Mach number, cone angle, and angle of attack. The normal force coefficient C_N, the axial force coefficient C_A and the center-of-pressure locations XCP/L, YCP/D, and XVCP/LV are then presented as a function of L/R_N and R_N/R_B (see Fig. 8). L is the axial position measured from the sphere tip. R_n is the nose tip or sphere radius, R_B is the radius of the base. D is the diameter of the base, LV is the axial length referenced to the virtual or sharp cone nosetip, XCP is the center-of-pressure location

referenced to the sphere tip, YCP is the center-of-pressure location referenced to the base and XVCP is the center-of-pressure location referenced to the virtual or sharp cone nosetip. The calculations are presented until $L/R_N \approx 200$.

MACH NO = 3.50 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0145	.9822	1.0955	-.0437	1.0077	1.0038
1.1525	.0161	.9482	.8817	.0670	.9883	.9831
1.4627	.0179	.9074	.7332	.1869	.9673	.9576
1.8577	.0198	.8596	.6340	.3151	.9449	.9269
2.4923	.0222	.7916	.5564	.4873	.9147	.8815
3.3075	.0241	.7169	.5090	.6734	.8822	.8294
4.1061	.0253	.6551	.4868	.8260	.8555	.7840
5.3069	.0265	.5787	.4758	1.0075	.8237	.7243
6.7549	.0273	.5066	.4739	1.1787	.7937	.6634
8.1078	.0278	.4538	.4780	1.3017	.7722	.6151
10.0645	.0282	.3952	.4882	1.4333	.7492	.5565
11.8604	.0284	.3545	.4988	1.5212	.7338	.5118
14.4204	.0286	.3113	.5136	1.6102	.7183	.4591
16.1378	.0287	.2892	.5229	1.6535	.7107	.4295
18.0013	.0288	.2696	.5322	1.6900	.7043	.4014
20.0198	.0288	.2524	.5415	1.7202	.6990	.3748
22.2026	.0289	.2374	.5507	1.7449	.6947	.3498
24.5598	.0290	.2243	.5596	1.7646	.6912	.3263
27.1020	.0291	.2129	.5682	1.7800	.6885	.3042
29.8409	.0292	.2030	.5764	1.7919	.6865	.2835
32.7884	.0293	.1945	.5842	1.8008	.6849	.2642
35.9576	.0295	.1871	.5916	1.8075	.6837	.2462
39.3623	.0296	.1807	.5985	1.8123	.6829	.2294
43.0172	.0298	.1751	.6049	1.8159	.6823	.2137
46.9379	.0299	.1704	.6108	1.8186	.6818	.1991
51.1413	.0301	.1663	.6162	1.8206	.6814	.1855
55.6450	.0302	.1627	.6211	1.8222	.6812	.1729
60.4679	.0304	.1596	.6256	1.8236	.6809	.1611
65.6302	.0306	.1570	.6297	1.8249	.6807	.1502
71.1534	.0307	.1547	.6334	1.8261	.6805	.1400
77.0602	.0309	.1528	.6368	1.8273	.6803	.1306
83.3749	.0310	.1511	.6399	1.8285	.6801	.1218
90.1234	.0311	.1496	.6426	1.8298	.6798	.1136
94.8770	.0312	.1488	.6443	1.8307	.6797	.1085
102.4106	.0314	.1476	.6466	1.8321	.6794	.1013
110.4555	.0315	.1466	.6487	1.8336	.6792	.0945
119.0443	.0316	.1458	.6506	1.8352	.6789	.0883
128.2118	.0317	.1451	.6524	1.8368	.6786	.0824
137.9948	.0318	.1444	.6539	1.8384	.6783	.0770
148.4330	.0319	.1439	.6553	1.8401	.6780	.0719
159.5694	.0320	.1434	.6566	1.8417	.6777	.0672
171.4460	.0321	.1430	.6577	1.8434	.6775	.0628
184.1137	.0322	.1426	.6588	1.8450	.6772	.0587
197.6228	.0322	.1423	.6598	1.8467	.6769	.0549
202.3221	.0323	.1422	.6600	1.8472	.6768	.0537

MACH NO = 5.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0142	.9355	1.0955	-.0437	1.0077	1.0038
1.1000	.0152	.9091	.9166	.0453	.9921	.9875
1.3410	.0164	.8773	.7765	.1450	.9746	.9674
1.7594	.0180	.8259	.6431	.2934	.9487	.9343
2.3143	.0195	.7648	.5557	.4595	.9196	.8938
3.0294	.0206	.6959	.4992	.6420	.8877	.8465
3.9243	.0214	.6228	.4655	.8326	.8543	.7939
5.0147	.0218	.5493	.4486	1.0202	.8215	.7380
6.3132	.0221	.4791	.4453	1.1923	.7914	.6809
7.8307	.0223	.4146	.4527	1.3382	.7659	.6244
9.5772	.0225	.3574	.4676	1.4534	.7457	.5701
11.5631	.0228	.3080	.4869	1.5385	.7308	.5187
13.7992	.0233	.2663	.5084	1.5974	.7205	.4709
16.2976	.0238	.2315	.5299	1.6354	.7138	.4272
18.3512	.0242	.2096	.5454	1.6542	.7106	.3965
20.5654	.0247	.1906	.5597	1.6673	.7083	.3682
22.9467	.0253	.1744	.5726	1.6770	.7066	.3420
25.5018	.0258	.1606	.5841	1.6848	.7052	.3177
28.2378	.0263	.1488	.5942	1.6918	.7040	.2953
31.1627	.0268	.1388	.6029	1.6987	.7029	.2745
34.2844	.0273	.1302	.6104	1.7057	.7015	.2554
37.6116	.0277	.1230	.6168	1.7130	.7003	.2377
41.1534	.0281	.1168	.6223	1.7205	.6990	.2214
44.9196	.0285	.1116	.6271	1.7281	.6976	.2063
50.3078	.0290	.1057	.6326	1.7381	.6959	.1881
54.6390	.0293	.1021	.6360	1.7455	.6946	.1755
59.2280	.0296	.0990	.6391	1.7526	.6933	.1640
64.0924	.0299	.0963	.6418	1.7595	.6921	.1533
69.2415	.0302	.0940	.6442	1.7660	.6910	.1434
74.6895	.0304	.0921	.6464	1.7723	.6899	.1342
80.4508	.0307	.0904	.6483	1.7784	.6888	.1257
86.5401	.0309	.0889	.6500	1.7841	.6878	.1178
92.9732	.0311	.0877	.6516	1.7896	.6869	.1105
99.7667	.0312	.0866	.6530	1.7947	.6860	.1037
106.9379	.0314	.0857	.6543	1.7997	.6851	.0974
117.1190	.0316	.0846	.6558	1.8058	.6840	.0896
125.2448	.0318	.0840	.6568	1.8102	.6833	.0842
133.8139	.0319	.0834	.6577	1.8143	.6825	.0792
142.8483	.0320	.0829	.6585	1.8182	.6819	.0745
152.3718	.0321	.0824	.6593	1.8218	.6812	.0702
162.4096	.0322	.0820	.6600	1.8253	.6806	.0661
172.9888	.0323	.0817	.6607	1.8286	.6800	.0623
184.1385	.0324	.0814	.6613	1.8317	.6795	.0587
195.8898	.0325	.0812	.6618	1.8346	.6790	.0554
208.0749	.0325	.0810	.6622	1.8365	.6786	.0533

MACH NO = 10.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0140	.8992	1.3955	-.0437	1.0077	1.0038
1.0958	.0148	.8740	.9197	.0435	.9924	.9879
1.3648	.0158	.8399	.7622	.1567	.9726	.9655
1.8111	.0170	.7852	.6224	.3182	.9443	.9304
2.3935	.0179	.7224	.5308	.4988	.9127	.8883
3.1523	.0183	.6513	.4689	.7022	.8771	.8388
4.0830	.0183	.5776	.4294	.9147	.8399	.7852
5.1968	.0179	.5051	.4059	1.1260	.8030	.7294
6.5029	.0174	.4365	.3960	1.3222	.7686	.6733
8.0064	.0168	.3742	.3976	1.4916	.7390	.6185
9.7088	.0162	.3191	.4082	1.6270	.7153	.5663
11.6098	.0158	.2716	.4254	1.7263	.6979	.5176
13.7081	.0155	.2313	.4471	1.7914	.6866	.4727
16.0030	.0153	.1975	.4714	1.8258	.6805	.4317
18.4947	.0154	.1693	.4972	1.8346	.6790	.3946
21.4944	.0155	.1514	.5168	1.8277	.6802	.3691
23.7397	.0159	.1311	.5425	1.8048	.6842	.3380
26.7988	.0163	.1143	.5671	1.7713	.6911	.3100
28.8422	.0168	.1036	.5845	1.7422	.6952	.2907
32.3147	.0176	.0915	.6058	1.7014	.7023	.2671
36.0563	.0185	.0814	.6248	1.6617	.7092	.2456
39.9578	.0193	.0750	.6373	1.6344	.7140	.2308
43.3489	.0204	.0676	.6517	1.6031	.7195	.2124
46.8053	.0213	.0629	.6608	1.5843	.7228	.1996
51.7634	.0226	.0576	.6705	1.5663	.7259	.1837
57.1575	.0239	.0531	.6777	1.5567	.7276	.1690
61.5116	.0248	.0503	.6816	1.5549	.7279	.1588
67.7556	.0260	.0471	.6851	1.5589	.7272	.1461
74.4668	.0271	.0446	.6869	1.5688	.7255	.1346
79.6681	.0279	.0430	.6874	1.5788	.7238	.1268
86.8039	.0287	.0413	.6875	1.5938	.7211	.1175
94.1635	.0295	.0399	.6870	1.6100	.7183	.1092
99.8281	.0301	.0391	.6864	1.6223	.7161	.1036
107.5746	.0306	.0381	.6854	1.6384	.7133	.0968
113.5336	.0310	.0376	.6846	1.6501	.7113	.0922
121.6916	.0315	.0369	.6835	1.6653	.7086	.0865
130.1208	.0320	.0363	.6824	1.6797	.7061	.0813
136.6467	.0322	.0360	.6816	1.6900	.7043	.0777
145.6618	.0326	.0356	.6805	1.7033	.7020	.0732
155.0866	.0329	.0353	.6794	1.7162	.6997	.0690
162.4565	.0331	.0350	.6786	1.7256	.6981	.0661
172.7231	.0333	.0348	.6775	1.7379	.6959	.0624
183.5317	.0335	.0346	.6763	1.7501	.6938	.0589
192.0169	.0337	.0344	.6754	1.7592	.6922	.0564
200.8442	.0338	.0343	.6745	1.7682	.6906	.0541

MACH NO = 15.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0139	.8922	1.0955	-.0437	1.0077	1.0038
1.1720	.0150	.8570	.8656	.0773	.9865	.9815
1.4544	.0159	.8209	.7244	.1920	.9664	.9582
1.9238	.0169	.7658	.5971	.3574	.9375	.9220
2.5427	.0176	.7011	.5111	.5458	.9045	.8781
3.3253	.0178	.6303	.4536	.7525	.8683	.8283
4.2733	.0176	.5580	.4158	.9675	.8307	.7751
5.3975	.0170	.4874	.3923	1.1810	.7933	.7202
6.7035	.0163	.4213	.3812	1.3801	.7585	.6654
8.1921	.0155	.3615	.3805	1.5539	.7281	.6124
9.8603	.0148	.3089	.3880	1.6959	.7033	.5621
11.7021	.0142	.2636	.4017	1.8042	.6843	.5154
14.2370	.0136	.2166	.4246	1.8947	.6685	.4626
16.4415	.0132	.1858	.4457	1.9352	.6614	.4247
19.4048	.0130	.1543	.4738	1.9529	.6583	.3826
21.9327	.0129	.1339	.4969	1.9461	.6595	.3527
25.2790	.0130	.1132	.5253	1.9182	.6644	.3197
28.8230	.0132	.0966	.5526	1.8756	.6718	.2909
31.7956	.0135	.0858	.5732	1.8350	.6789	.2704
35.6797	.0141	.0746	.5970	1.7803	.6885	.2477
39.7504	.0148	.0656	.6187	1.7245	.6982	.2276
43.1433	.0154	.0596	.6344	1.6810	.7059	.2132
47.5603	.0164	.0534	.6520	1.6299	.7148	.1970
52.1810	.0175	.0483	.6672	1.5839	.7229	.1824
56.0308	.0185	.0449	.6777	1.5516	.7285	.1719
61.0424	.0198	.0413	.6888	1.5179	.7344	.1598
65.2152	.0209	.0388	.6961	1.4964	.7382	.1510
70.6381	.0223	.0363	.7033	1.4767	.7416	.1409
76.2896	.0238	.0342	.7085	1.4646	.7437	.1317
80.9719	.0249	.0328	.7115	1.4600	.7445	.1250
87.0202	.0263	.0313	.7138	1.4598	.7446	.1172
93.2803	.0275	.0301	.7149	1.4647	.7437	.1102
98.4416	.0285	.0293	.7151	1.4716	.7425	.1049
105.1006	.0297	.0285	.7146	1.4831	.7405	.0989
112.0364	.0307	.0278	.7135	1.4972	.7380	.0933
117.8432	.0315	.0273	.7123	1.5101	.7358	.0891
125.5236	.0324	.0268	.7104	1.5278	.7327	.0840
132.0858	.0331	.0265	.7085	1.5435	.7299	.0802
140.9157	.0338	.0261	.7058	1.5651	.7261	.0755
150.5329	.0345	.0258	.7025	1.5896	.7219	.0710
158.6478	.0349	.0255	.6995	1.6108	.7181	.0676
169.2715	.0353	.0253	.6955	1.6388	.7133	.0636
180.4802	.0355	.0251	.6913	1.6673	.7083	.0599
189.9060	.0356	.0250	.6880	1.6899	.7043	.0570
202.3088	.0357	.0248	.6840	1.7173	.6995	.0537

MACH NO = 20.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0139	.8896	1.1955	-.0437	1.0077	1.0038
1.1695	.0149	.8547	.9671	.0763	.9866	.9817
1.4501	.0158	.8190	.7258	.1906	.9666	.9586
1.9179	.0168	.7641	.5976	.3558	.9377	.9225
2.5329	.0175	.6996	.5107	.5446	.9047	.8788
3.3079	.0177	.6294	.4525	.7510	.8686	.8294
4.2454	.0174	.5576	.4137	.9664	.8309	.7765
5.3552	.0168	.4876	.3899	1.1816	.7933	.7221
6.6422	.0160	.4219	.3761	1.3837	.7579	.6678
8.1057	.0152	.3625	.3734	1.5623	.7266	.6152
9.7413	.0144	.3101	.3789	1.7106	.7007	.5654
11.5513	.0137	.2650	.3904	1.8264	.6804	.5192
14.0084	.0130	.2182	.4136	1.9275	.6627	.4669
16.6975	.0124	.1805	.4346	1.9857	.6525	.4207
20.1905	.0120	.1453	.4658	2.0101	.6493	.3727
23.9422	.0118	.1185	.4976	1.9976	.6535	.3321
27.9205	.0119	.0981	.5283	1.9604	.6670	.2977
32.0951	.0121	.0824	.5571	1.9083	.6661	.2685
35.7036	.0124	.0720	.5793	1.8588	.6748	.2475
40.1677	.0129	.0622	.6038	1.7960	.6857	.2257
44.7578	.0135	.0544	.6258	1.7329	.6968	.2069
49.4571	.0144	.0482	.6455	1.6718	.7075	.1907
54.2535	.0153	.0433	.6629	1.6148	.7174	.1766
59.1384	.0165	.0393	.6780	1.5633	.7265	.1642
63.2719	.0175	.0366	.6889	1.5252	.7331	.1550
68.3026	.0188	.0339	.7001	1.4860	.7400	.1451
73.4040	.0202	.0317	.7093	1.4539	.7456	.1363
78.5692	.0216	.0299	.7166	1.4289	.7500	.1284
83.7907	.0231	.0284	.7223	1.4105	.7532	.1213
88.1814	.0243	.0274	.7260	1.3998	.7551	.1159
93.4982	.0258	.0264	.7291	1.3920	.7564	.1099
98.8794	.0272	.0255	.7312	1.3890	.7570	.1045
104.3557	.0286	.0248	.7324	1.3900	.7568	.0995
109.9808	.0299	.0242	.7327	1.3948	.7559	.0949
115.8286	.0312	.0237	.7323	1.4030	.7545	.0905
120.9331	.0322	.0234	.7314	1.4125	.7528	.0870
127.4120	.0333	.0230	.7297	1.4273	.7503	.0829
134.3646	.0343	.0227	.7272	1.4464	.7469	.0789
141.8654	.0353	.0224	.7237	1.4704	.7427	.0750
150.0072	.0360	.0222	.7193	1.4995	.7376	.0712
157.3712	.0365	.0220	.7149	1.5274	.7327	.0681
167.0210	.0369	.0218	.7090	1.5647	.7262	.0644
177.2939	.0371	.0217	.7028	1.6036	.7194	.0609
187.7624	.0372	.0215	.6968	1.6411	.7129	.0577
200.2644	.0372	.0214	.6903	1.6819	.7057	.0542

MACH NO = 25.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0139	.8886	1.0955	-.0437	1.0077	1.0038
1.1684	.0149	.8539	.9678	.0758	.9867	.9818
1.4481	.0158	.8182	.7263	.1900	.9668	.9587
1.9140	.0168	.7635	.5977	.3552	.9378	.9227
2.5284	.0174	.6931	.5134	.5441	.9048	.8791
3.3000	.0176	.6291	.4519	.7505	.8687	.8298
4.2326	.0173	.5576	.4127	.9661	.8310	.7772
5.3357	.0167	.4878	.3873	1.1819	.7932	.7230
6.6136	.0158	.4223	.3737	1.3854	.7576	.6689
8.0653	.0150	.3630	.3730	1.5663	.7259	.6165
9.6854	.0142	.3118	.3745	1.7176	.6995	.5670
11.4654	.0135	.2658	.3849	1.8370	.6786	.5210
13.3998	.0127	.2190	.4038	1.9433	.6600	.4690
17.0996	.0120	.1749	.4313	2.0157	.6473	.4145
21.1605	.0115	.1364	.4662	2.0409	.6429	.3612
25.5461	.0113	.1085	.5012	2.0217	.6462	.3173
30.1712	.0113	.0880	.5343	1.9755	.6543	.2812
34.9886	.0115	.0728	.5647	1.9148	.6650	.2514
39.9458	.0119	.0614	.5921	1.8471	.6768	.2267
44.9950	.0125	.0528	.6167	1.7772	.6890	.2061
50.1035	.0132	.0462	.6386	1.7083	.7011	.1887
55.2296	.0141	.0411	.6580	1.6430	.7125	.1740
60.3616	.0152	.0370	.6749	1.5831	.7230	.1614
65.4811	.0164	.0339	.6895	1.5298	.7323	.1505
70.9512	.0175	.0316	.7003	1.4897	.7393	.1423
74.9216	.0189	.0295	.7109	1.4496	.7464	.1338
79.9567	.0203	.0278	.7197	1.4165	.7521	.1264
84.9529	.0218	.0264	.7268	1.3901	.7558	.1198
89.9130	.0233	.0253	.7324	1.3697	.7613	.1139
94.8508	.0248	.0244	.7368	1.3548	.7629	.1085
99.7878	.0264	.0236	.7400	1.3446	.7647	.1037
104.7545	.0278	.0230	.7423	1.3389	.7657	.0992
109.8340	.0293	.0225	.7437	1.3372	.7660	.0950
115.2638	.0308	.0220	.7443	1.3396	.7656	.0910
120.5223	.0322	.0216	.7440	1.3462	.7645	.0873
125.4402	.0333	.0214	.7430	1.3555	.7628	.0841
131.5192	.0345	.0211	.7410	1.3713	.7611	.0805
138.0110	.0356	.0208	.7378	1.3927	.7563	.0770
145.0122	.0366	.0206	.7336	1.4200	.7515	.0735
152.6109	.0373	.0204	.7283	1.4531	.7457	.0701
160.8359	.0379	.0203	.7221	1.4910	.7391	.0667
169.8049	.0382	.0201	.7152	1.5332	.7317	.0634
179.7489	.0383	.0200	.7079	1.5771	.7240	.0602
188.9057	.0383	.0199	.7010	1.6188	.7167	.0573
201.2375	.0382	.0197	.6930	1.6680	.7091	.0540

MACH NO = 3[illegible] CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0133	.8879	1.0355	-.0437	1.0377	1.0038
1.1677	.0142	.8573	.9682	.0756	.9858	.9918
1.4470	.0153	.8177	.9267	.1896	.9668	.9588
1.9122	.0167	.7670	.8979	.3548	.9379	.9228
2.5258	.0174	.6988	.8734	.5437	.9049	.8793
3.2057	.0175	.6289	.8517	.7499	.8688	.8301
4.4838	.0171	.5330	.8346	1.0201	.8215	.7642
5.6269	.0164	.4711	.8219	1.2345	.7840	.7099
6.9403	.0155	.4071	.8105	1.4341	.7491	.6563
8.4236	.0147	.3496	.8085	1.6094	.7184	.6047
10.0811	.0139	.2993	.8040	1.7546	.5970	.5561
11.8875	.0132	.2561	.8051	1.8682	.6731	.5112
14.7468	.0124	.2113	.8041	1.9686	.6555	.4605
18.1240	.0117	.1632	.8060	2.0428	.6426	.3997
22.8498	.0111	.1239	.8751	2.0570	.6411	.3430
27.9160	.0103	.0965	.9131	2.0236	.6459	.2977
33.2375	.0110	.0772	.9479	1.9647	.6562	.2615
38.2310	.0112	.0648	.9753	1.9029	.6670	.2356
43.5951	.0117	.0543	.9835	1.8271	.6813	.2114
49.1856	.0124	.0466	.9895	1.7505	.6937	.1916
54.9601	.0131	.0414	.9879	1.6857	.7050	.1771
60.8741	.0141	.0368	.9674	1.6166	.7171	.1632
66.9048	.0153	.0333	.9842	1.5542	.7280	.1514
73.3367	.0166	.0310	.9995	1.4995	.7376	.1414
74.0145	.0179	.0286	.7092	1.4580	.7440	.1339
80.0436	.0192	.0260	.7195	1.4177	.7510	.1263
85.7659	.0207	.0255	.7279	1.3844	.7578	.1196
89.0019	.0223	.0243	.7348	1.3576	.7624	.1138
94.2356	.0237	.0235	.7397	1.3390	.7657	.1092
99.0414	.0253	.0227	.7441	1.3226	.7686	.1044
103.8370	.0269	.0221	.7475	1.3110	.7706	.1000
108.7697	.0283	.0216	.7496	1.3045	.7717	.0964
113.0075	.0299	.0212	.7512	1.3013	.7723	.0926
118.1047	.0315	.0208	.7518	1.3027	.7721	.0889
123.4446	.0331	.0205	.7515	1.3093	.7709	.0854
128.7646	.0342	.0202	.7502	1.3197	.7691	.0823
134.3047	.0355	.0200	.7477	1.3374	.7660	.0789
140.6622	.0366	.0198	.7440	1.3614	.7618	.0756
147.5248	.0376	.0196	.7392	1.3919	.7565	.0723
153.9121	.0382	.0195	.7340	1.4231	.7513	.0695
161.7871	.0387	.0193	.7273	1.4638	.7479	.0664
170.7732	.0390	.0192	.7197	1.5091	.7359	.0632
179.4140	.0391	.0191	.7127	1.5509	.7286	.0605
187.9134	.0391	.0189	.7048	1.5981	.7204	.0576
200.8962	.0388	.0188	.6949	1.6570	.7101	.0541

MACH NO = 3.50 CONE ANGLE = 6.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/L	XVCP/LV	RN/RB
.8955	.0144	.9849	1.1167	-.0526	1.0110	1.0055
1.1294	.0159	.9455	.8999	.0555	.9883	.9812
1.3485	.0172	.9113	.7820	.1411	.9703	.9596
1.8132	.0195	.8456	.6499	.2909	.9389	.9166
2.2891	.0213	.7864	.5846	.4167	.9124	.8764
2.8705	.0227	.7234	.5412	.5478	.8848	.8319
3.7625	.0242	.6430	.5084	.7136	.8500	.7717
4.6205	.0251	.5799	.4982	.8365	.8242	.7215
5.6176	.0259	.5202	.4953	.9508	.8001	.6708
7.0748	.0267	.4524	.4986	1.0781	.7734	.6083
8.4220	.0271	.4046	.5068	1.1631	.7555	.5600
9.9425	.0276	.3628	.5171	1.2341	.7406	.5140
12.1053	.0280	.3187	.5319	1.3051	.7257	.4603
13.5526	.0282	.2963	.5404	1.3395	.7184	.4301
15.1194	.0285	.2766	.5493	1.3687	.7123	.4017
17.4054	.0287	.2542	.5605	1.4007	.7056	.3663
19.2765	.0289	.2399	.5688	1.4202	.7015	.3417
21.2900	.0291	.2276	.5766	1.4363	.6981	.3187
24.2097	.0294	.2137	.5863	1.4535	.6945	.2903
26.5867	.0295	.2049	.5932	1.4637	.6923	.2707
29.1339	.0297	.1973	.5996	1.4720	.6906	.2524
32.8118	.0299	.1888	.6075	1.4808	.6887	.2299
35.7946	.0301	.1834	.6129	1.4861	.6876	.2145
38.9818	.0302	.1788	.6178	1.4905	.6867	.2001
42.3851	.0304	.1749	.6224	1.4942	.6859	.1867
47.2800	.0306	.1704	.6280	1.4994	.6850	.1704
51.2363	.0308	.1677	.6317	1.5011	.6845	.1591
55.4525	.0309	.1652	.6351	1.5035	.6839	.1486
61.5043	.0311	.1625	.6392	1.5064	.6833	.1358
66.3866	.0313	.1608	.6420	1.5084	.6829	.1269
71.5821	.0314	.1594	.6445	1.5103	.6825	.1187
79.0283	.0316	.1577	.6475	1.5127	.6820	.1086
85.0273	.0317	.1567	.6496	1.5144	.6817	.1017
91.4045	.0318	.1558	.6514	1.5161	.6813	.0952
100.5364	.0319	.1548	.6536	1.5183	.6808	.0872
107.8830	.0320	.1541	.6551	1.5199	.6805	.0817
115.6893	.0321	.1536	.6565	1.5215	.6802	.0766
123.9800	.0322	.1531	.6577	1.5230	.6798	.0718
139.8415	.0322	.1526	.6592	1.5251	.6794	.0659
145.3816	.0323	.1522	.6602	1.5265	.6791	.0618
155.5117	.0324	.1519	.6611	1.5280	.6788	.0580
169.9992	.0325	.1516	.6622	1.5298	.6784	.0533
181.6511	.0325	.1514	.6629	1.5312	.6781	.0500
194.0235	.0326	.1512	.6636	1.5326	.6778	.0470
222.6943	.0326	.1511	.6640	1.5334	.6777	.0450

MACH NO = 5.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0141	.9382	1.1167	-.0526	1.0110	1.0055
1.0780	.0151	.9078	.9362	.0339	.9929	.9865
1.3104	.0162	.8714	.7947	.1296	.9728	.9633
1.7134	.0177	.8137	.6603	.2693	.9434	.9255
2.2444	.0190	.7465	.5729	.4218	.9113	.8801
2.9243	.0201	.6727	.5169	.5849	.8771	.8280
3.5421	.0206	.6154	.4908	.7087	.8510	.7857
4.5199	.0210	.5398	.4719	.8678	.8176	.7270
5.6828	.0213	.4685	.4674	1.0104	.7876	.6677
7.0382	.0215	.4040	.4744	1.1278	.7629	.6097
8.5924	.0219	.3476	.4890	1.2173	.7441	.5545
9.8921	.0222	.3109	.5030	1.2670	.7337	.5154
11.8088	.0227	.2689	.5236	1.3134	.7239	.4670
13.9415	.0233	.2343	.5445	1.3422	.7179	.4227
15.6871	.0238	.2126	.5594	1.3557	.7150	.3923
18.2157	.0246	.1885	.5775	1.3671	.7126	.3552
20.2679	.0252	.1736	.5894	1.3730	.7114	.3300
23.2191	.0259	.1571	.6029	1.3798	.7100	.2993
25.5995	.0265	.1469	.6114	1.3849	.7089	.2785
29.0050	.0272	.1358	.6208	1.3926	.7073	.2532
31.7392	.0277	.1289	.6265	1.3990	.7059	.2361
35.6353	.0283	.1213	.6328	1.4081	.7040	.2152
38.7527	.0287	.1166	.6367	1.4153	.7025	.2011
43.1815	.0292	.1114	.6410	1.4249	.7005	.1839
46.7156	.0296	.1081	.6438	1.4320	.6990	.1721
51.7243	.0300	.1045	.6469	1.4412	.6970	.1578
55.7128	.0303	.1023	.6489	1.4478	.6957	.1480
61.3548	.0306	.0998	.6512	1.4561	.6939	.1361
65.8401	.0308	.0982	.6527	1.4620	.6927	.1279
72.1753	.0311	.0964	.6545	1.4693	.6911	.1178
77.2050	.0313	.0953	.6557	1.4745	.6900	.1109
84.3011	.0315	.0941	.6571	1.4809	.6887	.1025
89.9295	.0317	.0933	.6580	1.4854	.6878	.0966
95.8332	.0318	.0926	.6589	1.4897	.6869	.0911
104.1549	.0320	.0918	.6599	1.4949	.6858	.0844
110.7517	.0321	.0913	.6606	1.4986	.6850	.0797
120.0505	.0322	.0907	.6615	1.5032	.6840	.0740
127.4242	.0323	.0903	.6620	1.5065	.6833	.0700
137.8247	.0324	.0899	.6627	1.5106	.6825	.0650
146.0794	.0325	.0897	.6632	1.5135	.6819	.0615
157.7355	.0326	.0894	.6638	1.5172	.6811	.0572
166.9978	.0326	.0892	.6641	1.5198	.6805	.0542
180.0931	.0327	.0889	.6646	1.5232	.6798	.0504
190.5116	.0327	.0888	.6649	1.5257	.6793	.0478
201.4758	.0328	.0886	.6651	1.5280	.6788	.0453

MACH NO = 10.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	YCP/L	YCP/D	YVCP/LV	RN/RB
.0966	.0139	.0218	1.1167	-.0526	1.0110	1.0055
1.0719	.0146	.0229	.9401	.0317	.9933	.9871
1.8297	.0156	.0231	.7821	.1393	.9707	.9614
1.7552	.0167	.0233	.6414	.2910	.9390	.9218
2.7066	.0175	.0230	.5496	.4546	.9044	.8750
3.0065	.0178	.0228	.4891	.6314	.8673	.8221
3.8521	.0177	.0228	.4514	.8096	.8298	.7661
4.8525	.0174	.0227	.4298	.9809	.7938	.7090
5.8122	.0169	.0170	.4217	1.1346	.7615	.6526
7.3321	.0164	.0173	.4245	1.2626	.7346	.5994
8.8095	.0160	.0153	.4357	1.3619	.7139	.5476
10.0183	.0157	.0214	.4482	1.4151	.7025	.5119
11.7602	.0155	.0224	.4683	1.4635	.6924	.4681
14.1394	.0155	.0127	.4967	1.4910	.6866	.4190
16.1987	.0156	.0160	.5284	1.4923	.6963	.3942
18.0631	.0160	.0109	.5498	1.4753	.6899	.3456
21.3262	.0165	.0141	.5722	1.4520	.6948	.3183
23.8263	.0171	.0132	.5930	1.4240	.7007	.2937
27.1529	.0181	.0161	.6165	1.3868	.7085	.2664
29.9828	.0190	.0169	.6329	1.3581	.7145	.2468
33.7518	.0203	.0175	.6504	1.3263	.7212	.2248
36.9669	.0214	.0214	.6620	1.3055	.7256	.2090
40.3742	.0225	.0163	.6714	1.2896	.7289	.1944
44.9242	.0240	.0210	.6804	1.2771	.7315	.1779
48.8114	.0251	.0176	.6855	1.2729	.7324	.1658
53.9959	.0265	.0141	.6897	1.2741	.7322	.1521
58.4133	.0275	.0118	.6916	1.2796	.7310	.1420
63.0772	.0285	.0109	.6925	1.2880	.7292	.1328
69.2628	.0295	.0180	.6926	1.3016	.7264	.1222
74.5065	.0303	.0168	.6920	1.3140	.7238	.1145
79.9851	.0310	.0158	.6912	1.3269	.7211	.1074
87.0250	.0318	.0148	.6899	1.3430	.7177	.0995
92.8370	.0323	.0141	.6888	1.3556	.7151	.0938
100.3864	.0329	.0135	.6872	1.3710	.7118	.0873
106.6937	.0333	.0131	.6859	1.3831	.7093	.0825
113.2936	.0336	.0127	.6845	1.3954	.7067	.0781
121.9735	.0340	.0123	.6825	1.4111	.7034	.0729
129.2977	.0342	.0121	.6808	1.4241	.7006	.0690
138.9673	.0343	.0118	.6785	1.4406	.6972	.0645
147.1442	.0344	.0116	.6767	1.4536	.6944	.0611
155.7415	.0345	.0115	.6749	1.4661	.6918	.0579
167.1207	.0345	.0113	.6728	1.4809	.6887	.0542
176.7646	.0345	.0112	.6713	1.4916	.6864	.0513
189.5433	.0344	.0110	.6696	1.5037	.6839	.0480
200.3825	.0344	.0110	.6685	1.5122	.6821	.0455

MACH NO = 15.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/C	XVCP/LV	PN/PR
.8955	.0138	.8949	1.1167	-.0526	1.0119	1.0055
1.0677	.0146	.8667	.9433	.0299	.9937	.9875
1.3101	.0154	.8278	.7859	.1360	.9714	.9623
1.7391	.0164	.7688	.6433	.2863	.9398	.9232
2.2842	.0171	.7011	.5490	.4517	.9051	.8768
2.9667	.0174	.6282	.4866	.6282	.8679	.8249
3.7884	.0172	.5547	.4463	.8077	.8302	.7701
4.7567	.0167	.4837	.4213	.9828	.7934	.7141
5.8745	.0161	.4190	.4093	1.1432	.7597	.6588
7.1348	.0154	.3589	.4077	1.2808	.7308	.6058
8.5472	.0149	.3074	.4146	1.3909	.7076	.5559
9.6912	.0144	.2737	.4238	1.4549	.6942	.5211
11.7260	.0140	.2349	.4400	1.5158	.6812	.4783
13.9960	.0136	.1882	.4693	1.5662	.6708	.4217
16.8076	.0134	.1528	.5011	1.5750	.6689	.3736
20.0001	.0136	.1260	.5329	1.5558	.6738	.3331
23.2788	.0139	.1055	.5633	1.5135	.6816	.2988
26.7121	.0145	.0899	.5907	1.4745	.6901	.2697
29.6784	.0151	.0796	.6114	1.4345	.6985	.2488
33.7517	.0160	.0699	.6336	1.3869	.7085	.2270
37.1331	.0171	.0622	.6528	1.3422	.7179	.2092
41.0336	.0187	.0561	.6692	1.3020	.7263	.1918
45.0284	.0197	.0513	.6829	1.2675	.7336	.1775
49.1217	.0212	.0475	.6940	1.2394	.7395	.1640
53.2099	.0227	.0445	.7028	1.2180	.7440	.1538
56.9737	.0240	.0424	.7085	1.2052	.7466	.1454
61.2257	.0256	.0404	.7135	1.1955	.7487	.1363
65.6731	.0270	.0387	.7169	1.1913	.7496	.1282
70.2220	.0284	.0374	.7189	1.1918	.7495	.1208
74.8962	.0297	.0364	.7198	1.1961	.7486	.1140
79.7336	.0310	.0355	.7198	1.2038	.7470	.1077
84.8178	.0322	.0348	.7189	1.2144	.7447	.1019
89.2866	.0331	.0343	.7177	1.2255	.7424	.0972
95.0092	.0341	.0338	.7155	1.2416	.7390	.0919
101.2108	.0349	.0334	.7124	1.2614	.7349	.0867
107.9874	.0356	.0330	.7084	1.2852	.7298	.0816
115.4357	.0361	.0327	.7035	1.3131	.7240	.0767
123.6820	.0364	.0325	.6978	1.3444	.7174	.0719
131.1702	.0365	.0323	.6928	1.3720	.7116	.0681
140.5433	.0365	.0321	.6869	1.4038	.7049	.0638
150.3941	.0364	.0319	.6815	1.4335	.6987	.0599
160.8546	.0362	.0318	.6766	1.4605	.6930	.0562
172.0995	.0360	.0317	.6724	1.4844	.6880	.0527
184.3394	.0358	.0316	.6689	1.5052	.6836	.0493
200.1936	.0355	.0315	.6657	1.5251	.6794	.0456

MACH NO = 20.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0138	.8922	1.1167	-.0526	1.0110	1.0055
1.0662	.0145	.8643	.9445	.0292	.9939	.9877
1.3166	.0153	.8258	.7873	.1348	.9717	.9627
1.7332	.0163	.7671	.6441	.2849	.9401	.9237
2.2752	.0170	.6997	.5489	.4504	.9053	.8775
2.9510	.0172	.6273	.4858	.6268	.8683	.8261
3.7637	.0170	.5544	.4446	.8065	.8305	.7716
4.7198	.0165	.4839	.4184	.9828	.7934	.7161
5.8214	.0158	.4185	.4048	1.1456	.7592	.6613
7.0658	.0151	.3598	.4015	1.2868	.7295	.6086
8.4464	.0144	.3086	.4065	1.4017	.7054	.5592
9.9539	.0138	.2646	.4176	1.4890	.6870	.5137
11.5776	.0134	.2275	.4329	1.5506	.6741	.4723
15.1285	.0128	.1706	.4701	1.6094	.6617	.4015
19.0137	.0126	.1312	.5101	1.6067	.6622	.3450
22.6262	.0127	.1066	.5437	1.5744	.6691	.3050
26.9350	.0131	.0864	.5787	1.5205	.6854	.2680
31.3753	.0139	.0720	.6093	1.4602	.6931	.2382
35.7254	.0147	.0627	.6325	1.4071	.7042	.2168
39.8741	.0158	.0547	.6554	1.3492	.7164	.1964
44.4285	.0172	.0487	.6749	1.2967	.7274	.1795
49.4014	.0186	.0447	.6891	1.2564	.7359	.1670
52.9166	.0202	.0411	.7026	1.2176	.7440	.1547
57.4004	.0220	.0384	.7133	1.1867	.7505	.1442
61.2993	.0235	.0366	.7216	1.1660	.7549	.1362
65.7365	.0254	.0349	.7270	1.1489	.7585	.1280
70.1747	.0272	.0336	.7316	1.1381	.7608	.1208
74.0883	.0287	.0326	.7342	1.1332	.7618	.1151
78.6446	.0304	.0318	.7360	1.1324	.7620	.1091
83.3571	.0321	.0311	.7366	1.1364	.7611	.1035
87.6722	.0334	.0306	.7360	1.1440	.7595	.0989
92.0025	.0349	.0302	.7342	1.1578	.7566	.0938
98.5383	.0363	.0298	.7309	1.1778	.7524	.0888
103.8658	.0369	.0295	.7268	1.2006	.7476	.0846
110.5075	.0376	.0293	.7227	1.2330	.7408	.0799
117.7990	.0381	.0290	.7134	1.2710	.7328	.0753
124.7819	.0382	.0289	.7052	1.3079	.7251	.0713
133.0028	.0381	.0287	.6980	1.3497	.7163	.0672
141.3251	.0379	.0286	.6904	1.3887	.7081	.0635
148.7477	.0377	.0285	.6843	1.4199	.7015	.0605
157.4744	.0373	.0284	.6792	1.4519	.6948	.0573
166.5651	.0370	.0283	.6739	1.4798	.6889	.0543
174.9176	.0367	.0282	.6691	1.5008	.6845	.0519
185.0270	.0364	.0282	.6655	1.5208	.6803	.0491
200.2056	.0360	.0281	.6621	1.5417	.6759	.0456

MACH NO = 25.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 1.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0138	.8912	1.1167	-.0526	1.0110	1.0055
1.0665	.0145	.8634	.9451	.0289	.9939	.9878
1.3150	.0153	.8250	.7880	.1342	.9718	.9628
1.7305	.0163	.7665	.6444	.2843	.9402	.9240
2.2712	.0169	.6992	.5488	.4498	.9054	.8779
2.9439	.0171	.6271	.4853	.6262	.8684	.8266
3.7524	.0169	.5544	.4437	.8062	.8305	.7723
4.7027	.0164	.4841	.4170	.9829	.7934	.7170
5.7967	.0157	.4189	.4027	1.1468	.7589	.6624
7.0317	.0149	.3604	.3986	1.2897	.7289	.6100
8.3992	.0142	.3092	.4027	1.4068	.7043	.5608
9.8907	.0136	.2653	.4128	1.4968	.6854	.5155
11.4944	.0131	.2283	.4272	1.5613	.6718	.4743
15.9176	.0123	.1601	.4723	1.6321	.6569	.3886
20.2916	.0121	.1204	.5154	1.6208	.6593	.3297
24.9180	.0123	.0938	.5555	1.5735	.6692	.2841
29.6923	.0128	.0757	.5905	1.5117	.6822	.2487
34.5225	.0136	.0631	.6206	1.4462	.6960	.2208
39.3346	.0146	.0541	.6462	1.3819	.7095	.1986
44.5967	.0161	.0470	.6704	1.3155	.7235	.1790
49.2225	.0175	.0424	.6884	1.2626	.7346	.1646
53.7321	.0192	.0389	.7033	1.2173	.7441	.152.
58.1192	.0209	.0363	.7154	1.1800	.7520	.1427
62.3996	.0228	.0343	.7250	1.1502	.7582	.1341
66.5910	.0246	.0328	.7325	1.1273	.7630	.1266
70.7237	.0265	.0316	.7382	1.1107	.7665	.1200
75.2984	.0286	.0305	.7427	1.0989	.7690	.1134
79.4585	.0304	.0298	.7453	1.0938	.7701	.1081
83.7240	.0322	.0292	.7465	1.0939	.7700	.1031
88.1644	.0338	.0287	.7464	1.0996	.7689	.0984
92.8559	.0354	.0283	.7449	1.1112	.7664	.0938
97.8579	.0367	.0280	.7418	1.1294	.7626	.0894
103.8721	.0380	.0278	.7366	1.1577	.7566	.0846
109.7290	.0387	.0275	.7303	1.1903	.7498	.0804
116.0875	.0392	.0274	.7226	1.2289	.7417	.0763
122.9562	.0394	.0272	.7140	1.2721	.7326	.0723
130.3377	.0392	.0271	.7048	1.3178	.7230	.0685
137.8197	.0390	.0270	.6960	1.3615	.7138	.0650
145.3021	.0386	.0269	.6881	1.4013	.7054	.0618
153.6890	.0381	.0268	.6803	1.4404	.6972	.0586
161.3734	.0376	.0267	.6744	1.4708	.6908	.0560
169.2572	.0372	.0267	.6695	1.4967	.6854	.0535
177.4112	.0369	.0266	.6655	1.5181	.6809	.0512
185.9114	.0365	.0265	.6624	1.5354	.6773	.0489
200.0240	.0361	.0265	.6592	1.5545	.6732	.0456

MACH NO = 30.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0138	.8906	1.1167	-.0526	1.0110	1.0055
1.0651	.0145	.8629	.9454	.0287	.9940	.9878
1.3140	.0153	.8246	.7884	.1339	.9719	.9629
1.7289	.0163	.7661	.6446	.2839	.9403	.9241
2.2688	.0169	.6989	.5487	.4495	.9055	.8781
2.9398	.0171	.6269	.4851	.6258	.8685	.8269
3.7458	.0169	.5543	.4433	.8058	.8306	.7727
4.6930	.0163	.4842	.4163	.9828	.7934	.7175
5.7827	.0156	.4191	.4016	1.1473	.7588	.6650
7.0118	.0149	.3606	.3970	1.2911	.7286	.6107
8.3726	.0141	.3095	.4006	1.4095	.7037	.5617
9.8551	.0135	.2657	.4102	1.5010	.6845	.5165
11.4476	.0130	.2287	.4240	1.5672	.6706	.4754
16.2967	.0121	.1553	.4726	1.6444	.6543	.3827
21.1491	.0119	.1140	.5192	1.6277	.6578	.3202
26.2575	.0121	.0874	.5615	1.5730	.6693	.2732
32.0000	.0127	.0694	.6007	1.4982	.6851	.2345
37.2019	.0136	.0570	.6308	1.4278	.6999	.2079
42.2950	.0148	.0490	.6563	1.3597	.7142	.1871
47.2264	.0161	.0434	.6778	1.2972	.7273	.1705
52.4360	.0170	.0390	.6974	1.2374	.7399	.1560
56.9745	.0196	.0360	.7118	1.1918	.7495	.1452
61.3390	.0215	.0338	.7234	1.1544	.7573	.1361
65.5538	.0234	.0321	.7327	1.1246	.7636	.1284
70.0596	.0256	.0307	.7405	1.0998	.7698	.1210
74.0861	.0275	.0298	.7459	1.0837	.7722	.1151
78.1021	.0295	.0290	.7496	1.0734	.7744	.1098
82.5809	.0316	.0283	.7521	1.0684	.7754	.1044
86.7810	.0334	.0279	.7529	1.0700	.7751	.0998
91.1793	.0352	.0275	.7521	1.0779	.7734	.0954
95.8322	.0367	.0272	.7498	1.0927	.7703	.0911
101.7210	.0382	.0269	.7453	1.1172	.7651	.0866
106.6712	.0391	.0267	.7395	1.1470	.7589	.0826
112.4152	.0398	.0265	.7323	1.1835	.7512	.0786
118.6072	.0401	.0264	.7238	1.2256	.7424	.0748
125.8885	.0401	.0262	.7135	1.2763	.7317	.0708
132.9899	.0398	.0261	.7037	1.3244	.7216	.0672
140.2603	.0394	.0260	.6943	1.3705	.7119	.0639
148.2258	.0388	.0259	.6852	1.4158	.7024	.0607
156.4867	.0383	.0259	.6781	1.4516	.6949	.0580
162.8254	.0378	.0258	.6720	1.4822	.6884	.0555
170.7012	.0373	.0257	.6671	1.5079	.6830	.0532
178.7519	.0369	.0257	.6629	1.5305	.6783	.0508
196.7261	.0366	.0256	.6600	1.5465	.6749	.0487
208.1375	.0361	.0255	.6572	1.5637	.6713	.0456

MACH NO = 3.50 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP L	YCP/D	XVCP/LV	RN/RB
.8781	.0143	.9882	1.1388	-.0614	1.0151	1.0075
1.1065	.0158	.9438	.9187	.0440	.9892	.9798
1.3136	.0170	.9056	.7994	.1264	.9690	.9553
1.6692	.0187	.8488	.6875	.2393	.9412	.9177
2.1044	.0204	.7865	.6146	.3548	.9129	.8748
2.6363	.0219	.7208	.5665	.4728	.8839	.8275
3.4515	.0232	.6377	.5305	.6192	.8479	.7642
4.2347	.0242	.5734	.5184	.7259	.8217	.7119
5.1431	.0250	.5133	.5149	.8227	.7980	.6595
6.1845	.0256	.4596	.5170	.9084	.7769	.6082
7.3671	.0262	.4102	.5239	.9801	.7593	.5589
8.6992	.0268	.3681	.5337	1.0386	.7450	.5121
10.5882	.0274	.3240	.5476	1.0962	.7308	.4577
11.8482	.0277	.3018	.5562	1.1238	.7240	.4274
13.6850	.0281	.2765	.5673	1.1543	.7165	.3899
15.7112	.0285	.2556	.5778	1.1788	.7105	.3554
17.3623	.0288	.2423	.5851	1.1940	.7068	.3315
19.7494	.0291	.2275	.5942	1.2108	.7027	.3021
21.6865	.0294	.2182	.6004	1.2213	.7001	.2819
24.4765	.0297	.2077	.6081	1.2329	.6972	.2571
27.5179	.0300	.1992	.6150	1.2423	.6949	.2345
29.9739	.0301	.1979	.6198	1.2483	.6934	.2191
33.4964	.0304	.1879	.6255	1.2552	.6918	.2001
37.3197	.0306	.1830	.6306	1.2609	.6904	.1829
40.3970	.0308	.1799	.6340	1.2647	.6894	.1711
44.7972	.0310	.1765	.6382	1.2691	.6884	.1566
49.5588	.0312	.1737	.6420	1.2729	.6874	.1435
53.3823	.0313	.1719	.6445	1.2755	.6868	.1344
58.8383	.0315	.1699	.6476	1.2786	.6860	.1233
63.2146	.0316	.1687	.6496	1.2808	.6855	.1157
69.4534	.0317	.1673	.6521	1.2834	.6848	.1062
76.1844	.0319	.1661	.6544	1.2859	.6842	.0977
81.5772	.0319	.1654	.6559	1.2877	.6838	.0917
89.2579	.0321	.1645	.6577	1.2899	.6832	.0844
97.5378	.0322	.1638	.6593	1.2920	.6827	.0778
104.1684	.0322	.1634	.6604	1.2936	.6823	.0731
113.6098	.0323	.1629	.6617	1.2956	.6819	.0674
123.7874	.0324	.1625	.6628	1.2975	.6814	.0622
131.9392	.0324	.1622	.6636	1.2988	.6810	.0585
143.5508	.0325	.1619	.6646	1.3006	.6806	.0540
152.8545	.0325	.1618	.6652	1.3019	.6803	.0509
166.1126	.0325	.1615	.6660	1.3036	.6799	.0470
180.4232	.0326	.1614	.6667	1.3052	.6795	.0434
191.8920	.0326	.1612	.6672	1.3063	.6792	.0409
204.0430	.0326	.1611	.6676	1.3073	.6790	.0386

MACH NO = 5.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0140	.9414	1.1388	-.0614	1.0151	1.0075
1.0551	.0150	.9072	.9565	.0226	.9944	.9859
1.2801	.0160	.8668	.8135	.1146	.9719	.9598
1.5607	.0171	.8203	.7068	.2126	.9478	.9291
2.0369	.0184	.7503	.6086	.3512	.9137	.8812
2.6473	.0194	.6738	.5460	.4968	.8780	.8266
3.2024	.0199	.6148	.5164	.6060	.8512	.7825
4.0811	.0204	.5376	.4945	.7443	.8172	.7216
5.1250	.0207	.4654	.4882	.8662	.7873	.6605
6.0193	.0209	.4161	.4915	.9424	.7686	.6158
7.3631	.0212	.3581	.5035	1.0218	.7491	.5590
8.4860	.0216	.3203	.5162	1.0654	.7384	.5190
10.1385	.0221	.2775	.5356	1.1055	.7285	.4696
11.9709	.0228	.2423	.5556	1.1298	.7226	.4247
13.9864	.0236	.2138	.5746	1.1434	.7192	.3843
16.1884	.0244	.1909	.5915	1.1511	.7173	.3481
18.5812	.0253	.1726	.6059	1.1563	.7161	.3158
21.1696	.0261	.1580	.6178	1.1612	.7148	.2870
23.9591	.0269	.1464	.6272	1.1671	.7134	.2613
26.1869	.0275	.1392	.6329	1.1723	.7121	.2439
29.3436	.0281	.1314	.6391	1.1800	.7102	.2228
32.7201	.0287	.1251	.6439	1.1886	.7081	.2040
36.3243	.0293	.1201	.6476	1.1974	.7060	.1871
40.1652	.0297	.1160	.6507	1.2062	.7038	.1719
44.2520	.0302	.1127	.6531	1.2147	.7017	.1583
48.5951	.0305	.1100	.6552	1.2229	.6997	.1460
53.2055	.0309	.1078	.6569	1.2305	.6978	.1348
58.0953	.0311	.1060	.6584	1.2376	.6961	.1247
63.2779	.0314	.1046	.6596	1.2443	.6944	.1155
68.7684	.0316	.1033	.6608	1.2504	.6929	.1072
73.0988	.0318	.1026	.6615	1.2547	.6919	.1014
79.1718	.0320	.1017	.6624	1.2601	.6906	.0943
85.6083	.0321	.1010	.6632	1.2651	.6893	.0877
92.4354	.0322	.1004	.6639	1.2698	.6882	.0817
99.6844	.0324	.0998	.6645	1.2742	.6871	.0762
107.3907	.0325	.0994	.6650	1.2784	.6861	.0711
115.5929	.0326	.0990	.6655	1.2824	.6851	.0663
124.3323	.0326	.0987	.6659	1.2862	.6842	.0619
133.6528	.0327	.0985	.6662	1.2898	.6833	.0578
143.6000	.0327	.0983	.6665	1.2931	.6824	.0540
154.2216	.0328	.0981	.6668	1.2962	.6817	.0505
162.6604	.0328	.0979	.6671	1.2983	.6812	.0479
174.5839	.0328	.0978	.6674	1.3007	.6806	.0448
187.3254	.0328	.0977	.6678	1.3028	.6801	.0419
200.9428	.0328	.0976	.6681	1.3046	.6796	.0391

MACH NO = 10.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0138	.9050	1.1388	-.0614	1.0151	1.0075
1.0480	.0145	.8726	.9615	.0199	.9951	.9868
1.2951	.0154	.8288	.8027	.1224	.9699	.9581
1.7006	.0164	.7639	.6612	.2635	.9353	.9145
2.2211	.0171	.6913	.5692	.4134	.8985	.8640
2.8631	.0174	.6155	.5099	.5675	.8606	.8089
3.4262	.0174	.5592	.4808	.6813	.8327	.7660
4.2903	.0171	.4871	.4566	.8258	.7972	.7085
5.2860	.0167	.4209	.4464	.9540	.7657	.6520
6.4118	.0163	.3619	.4472	1.0599	.7397	.5981
7.6631	.0160	.3107	.4566	1.1405	.7199	.5478
8.6802	.0158	.2774	.4677	1.1845	.7091	.5127
10.1351	.0157	.2392	.4859	1.2235	.6995	.4697
12.5145	.0157	.1935	.5174	1.2471	.6937	.4130
14.6635	.0161	.1642	.5446	1.2434	.6947	.3724
16.9526	.0166	.1410	.5710	1.2260	.6989	.3371
19.3755	.0173	.1227	.5954	1.2008	.7051	.3064
21.9295	.0182	.1082	.6174	1.1726	.7120	.2795
25.1687	.0195	.0947	.6400	1.1396	.7202	.2516
28.0182	.0207	.0859	.6557	1.1153	.7261	.2312
31.0108	.0220	.0788	.6684	1.0956	.7310	.2131
34.1534	.0234	.0731	.6785	1.0811	.7345	.1969
37.4526	.0247	.0685	.6861	1.0720	.7368	.1824
41.6252	.0263	.0642	.6924	1.0679	.7379	.1668
45.2884	.0276	.0613	.6956	1.0695	.7374	.1551
49.1234	.0287	.0590	.6974	1.0747	.7361	.1446
53.1399	.0298	.0572	.6980	1.0828	.7341	.1350
57.3554	.0307	.0557	.6979	1.0928	.7316	.1261
62.7252	.0318	.0543	.6970	1.1066	.7282	.1165
67.5253	.0325	.0534	.6958	1.1193	.7251	.1090
72.7000	.0332	.0526	.6942	1.1328	.7218	.1019
78.3096	.0338	.0519	.6923	1.1475	.7182	.0952
85.4648	.0344	.0513	.6894	1.1663	.7135	.0879
91.7859	.0347	.0509	.6867	1.1831	.7095	.0823
98.4093	.0349	.0506	.6836	1.2007	.7051	.0771
105.5492	.0349	.0503	.6804	1.2183	.7008	.0722
113.0610	.0350	.0501	.6773	1.2352	.6967	.0677
122.6964	.0349	.0499	.6740	1.2539	.6921	.0627
131.2874	.0348	.0497	.6716	1.2676	.6887	.0588
140.4298	.0347	.0496	.6697	1.2795	.6858	.0552
150.1629	.0346	.0495	.6682	1.2894	.6834	.0518
160.5267	.0345	.0494	.6670	1.2976	.6814	.0486
173.8543	.0344	.0493	.6662	1.3052	.6795	.0450
185.7543	.0343	.0492	.6658	1.3100	.6783	.0422
201.0569	.0343	.0491	.6657	1.3143	.6773	.0391

MACH NO = 15.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	YYCP/LV	RN/RB
.8781	.0133	.8980	1.1788	-.0614	1.0151	1.0075
1.0440	.0144	.8665	.9647	.0102	.9955	.9872
1.2858	.0152	.8236	.8066	.1133	.9707	.9591
1.6851	.0162	.7596	.6632	.2600	.9362	.9161
2.1967	.0168	.6880	.5600	.4100	.8993	.8663
2.8214	.0170	.6136	.5078	.5640	.8615	.8123
3.5656	.0168	.5401	.4687	.7162	.8241	.7561
4.4326	.0164	.4704	.4451	.8607	.7886	.6998
5.4226	.0158	.4066	.4341	.9895	.7570	.6449
6.5312	.0152	.3500	.4334	1.0970	.7306	.5929
7.7511	.0148	.3011	.4406	1.1856	.7101	.5445
8.7334	.0145	.2693	.4498	1.2276	.6985	.5110
10.1248	.0142	.2328	.4656	1.2714	.6878	.4699
13.1490	.0139	.1769	.5034	1.3064	.6792	.4001
16.0135	.0141	.1424	.5382	1.2969	.6815	.3508
19.0282	.0145	.1173	.5713	1.2664	.6890	.3105
22.1663	.0153	.0989	.6011	1.2250	.6989	.2773
25.3950	.0162	.0852	.6270	1.1833	.7094	.2498
28.6896	.0174	.0750	.6490	1.1423	.7195	.2269
32.0303	.0188	.0672	.6676	1.1051	.7286	.2076
35.4028	.0203	.0613	.6829	1.0729	.7365	.1911
38.7976	.0219	.0568	.6953	1.0464	.7430	.1770
42.2989	.0238	.0528	.7063	1.0234	.7487	.1632
46.1321	.0255	.0502	.7134	1.0095	.7521	.1527
49.5909	.0271	.0481	.7185	1.0010	.7542	.1434
53.0912	.0287	.0465	.7219	.9972	.7551	.1351
56.6603	.0303	.0452	.7239	.9976	.7550	.1275
60.3384	.0317	.0442	.7246	1.0017	.7540	.1206
64.1768	.0330	.0434	.7243	1.0094	.7521	.1141
68.2359	.0342	.0427	.7229	1.0207	.7493	.1080
73.2306	.0354	.0421	.7198	1.0386	.7449	.1013
77.9947	.0362	.0417	.7159	1.0592	.7399	.0956
82.1999	.0368	.0414	.7106	1.0844	.7337	.0901
88.0358	.0372	.0411	.7043	1.1140	.7264	.0847
95.2596	.0373	.0409	.6971	1.1456	.7184	.0795
101.9372	.0372	.0407	.6900	1.1792	.7104	.0746
108.0096	.0369	.0405	.6833	1.2096	.7029	.0701
116.2786	.0366	.0403	.6773	1.2373	.6962	.0660
125.3407	.0363	.0402	.6716	1.2646	.6894	.0614
133.9846	.0359	.0401	.6676	1.2844	.6846	.0577
143.4614	.0356	.0400	.6647	1.3001	.6807	.0541
153.9604	.0354	.0399	.6628	1.3117	.6779	.0505
165.7116	.0352	.0398	.6618	1.3198	.6759	.0471
178.9967	.0350	.0397	.6615	1.3249	.6746	.0437
200.7035	.0349	.0395	.6621	1.3283	.6738	.0392

MACH NO = 20.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0137	.8954	1.1388	-.0614	1.0151	1.0075
1.0426	.0144	.8642	.8659	.0176	.9957	.9874
1.2824	.0152	.8217	.8081	.1181	.9710	.9595
1.6794	.0161	.7579	.7640	.2586	.9365	.9167
2.1865	.0167	.6869	.7031	.4085	.8997	.8672
2.8052	.0168	.6130	.6572	.5624	.8619	.8136
3.5407	.0166	.5400	.6171	.7149	.8244	.7579
4.3965	.0162	.4707	.5824	.8605	.7887	.7020
5.3720	.0156	.4072	.5530	.9914	.7565	.6475
6.4621	.0150	.3509	.5277	1.1019	.7294	.5959
7.6586	.0144	.3022	.5072	1.1892	.7080	.5479
8.9598	.0141	.2714	.4912	1.2594	.6956	.5146
9.9774	.0137	.2342	.4855	1.2876	.6878	.4740
13.6870	.0132	.1671	.5011	1.3335	.6725	.3909
17.3007	.0134	.1290	.5423	1.3159	.6768	.3323
21.0830	.0139	.1018	.5818	1.2722	.6876	.2879
25.3831	.0149	.0824	.6172	1.2142	.7018	.2499
29.2670	.0160	.0704	.6443	1.1624	.7146	.2233
33.5381	.0176	.0610	.6632	1.1090	.7277	.1999
37.7063	.0193	.0551	.6875	1.0666	.7381	.1830
42.0886	.0211	.0507	.7024	1.0308	.7469	.1690
46.9816	.0232	.0472	.7153	.9992	.7546	.1561
48.5000	.0252	.0448	.7242	.9779	.7599	.1462
51.9712	.0272	.0430	.7318	.9628	.7636	.1376
55.8139	.0293	.0415	.7358	.9528	.7661	.1292
59.7071	.0312	.0405	.7385	.9494	.7669	.1224
62.8877	.0331	.0397	.7397	.9511	.7664	.1162
67.0411	.0348	.0390	.7391	.9593	.7644	.1097
71.0101	.0362	.0385	.7369	.9727	.7611	.1041
75.7587	.0375	.0381	.7324	.9950	.7556	.0982
80.4071	.0383	.0378	.7265	1.0220	.7490	.0930
85.4471	.0388	.0376	.7191	1.0549	.7409	.0879
91.5570	.0393	.0374	.7094	1.0971	.7306	.0825
97.4870	.0398	.0372	.7010	1.1377	.7216	.0778
103.6110	.0394	.0371	.6909	1.1770	.7110	.0735
110.5647	.0393	.0370	.6818	1.2164	.7013	.0692
117.0343	.0394	.0368	.6749	1.2472	.6937	.0656
123.7840	.0389	.0368	.6632	1.2732	.6874	.0622
131.7212	.0384	.0366	.6643	1.2961	.6817	.0586
139.3666	.0381	.0366	.6613	1.3116	.6779	.0556
148.5744	.0377	.0365	.6592	1.3235	.6750	.0523
157.6715	.0375	.0364	.6584	1.3301	.6734	.0494
167.7323	.0373	.0363	.6585	1.3335	.6725	.0466
180.3408	.0372	.0362	.6592	1.3345	.6723	.0434
201.3166	.0371	.0361	.6610	1.3328	.6727	.0391

MACH NO = 25.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0137	.8944	1.1388	-.0614	1.0151	1.0075
1.0419	.0144	.8633	.9664	.0173	.9958	.9875
1.2808	.0152	.8209	.8087	.1176	.9711	.9597
1.5768	.0160	.7574	.6644	.2580	.9366	.9169
2.1819	.0166	.6865	.5692	.4077	.8999	.8676
2.7976	.0168	.6129	.5069	.5616	.8621	.8142
3.5291	.0165	.5401	.4664	.7143	.8246	.7587
4.3796	.0161	.4710	.4411	.8694	.7887	.7030
5.3481	.0154	.4077	.4280	.9923	.7563	.6488
6.4294	.0148	.3515	.4249	1.1042	.7298	.5973
7.6150	.0142	.3028	.4297	1.1933	.7070	.5495
8.9661	.0139	.2711	.4371	1.2451	.6942	.5164
9.9078	.0135	.2346	.4505	1.2954	.6819	.4759
14.3302	.0129	.1575	.5027	1.3476	.6691	.3782
18.7314	.0130	.1156	.5519	1.3178	.6764	.3140
23.2856	.0137	.0896	.5946	1.2607	.6904	.2671
27.8511	.0148	.0729	.6295	1.1996	.7057	.2323
32.3219	.0162	.0619	.6580	1.1388	.7204	.2060
36.6332	.0180	.0544	.6814	1.0842	.7338	.1858
40.7602	.0199	.0492	.7003	1.0370	.7453	.1698
44.7072	.0221	.0456	.7153	.9983	.7548	.1569
48.4993	.0243	.0429	.7269	.9681	.7623	.1462
52.5016	.0268	.0409	.7363	.9441	.7681	.1364
56.1031	.0291	.0395	.7424	.9298	.7717	.1286
59.6993	.0313	.0385	.7462	.9221	.7736	.1217
63.3559	.0334	.0377	.7481	.9211	.7738	.1154
67.1463	.0353	.0371	.7479	.9270	.7724	.1095
71.1507	.0370	.0367	.7457	.9403	.7691	.1039
75.4333	.0384	.0363	.7413	.9615	.7639	.0986
80.0481	.0394	.0361	.7348	.9908	.7567	.0933
85.5021	.0400	.0358	.7254	1.0312	.7468	.0878
90.8894	.0400	.0357	.7153	1.0743	.7362	.0830
96.6706	.0397	.0356	.7042	1.1208	.7248	.0784
102.7425	.0392	.0354	.6933	1.1671	.7134	.0741
108.9450	.0385	.0353	.6833	1.2096	.7030	.0701
115.2469	.0378	.0352	.6748	1.2462	.6940	.0665
121.7177	.0372	.0352	.6680	1.2764	.6866	.0632
128.4322	.0367	.0351	.6628	1.3002	.6807	.0600
136.1309	.0362	.0350	.6589	1.3192	.6761	.0569
143.6311	.0359	.0349	.6568	1.3307	.6732	.0540
151.6709	.0356	.0348	.6560	1.3374	.6716	.0513
160.3847	.0355	.0347	.6561	1.3402	.6709	.0486
169.9473	.0354	.0347	.6569	1.3401	.6709	.0460
180.5756	.0353	.0346	.6583	1.3381	.6714	.0434
201.0195	.0353	.0345	.6610	1.3329	.6727	.0391

MACH NO = 30.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8791	.0137	.8937	1.1388	-.0614	1.0151	1.0075
1.0415	.0144	.8627	.9667	.0171	.9958	.9876
1.2809	.0152	.8204	.8091	.1172	.9712	.9598
1.6757	.0160	.7570	.6646	.2576	.9367	.9171
2.1793	.0166	.6862	.5692	.4073	.9000	.8678
2.7933	.0167	.6128	.5067	.5612	.8622	.8145
3.5226	.0165	.5401	.4660	.7140	.8247	.7592
4.3792	.0160	.4711	.4404	.8603	.7887	.7036
5.3349	.0154	.4079	.4270	.9927	.7562	.6494
6.4113	.0147	.3518	.4235	1.1054	.7285	.5981
7.5907	.0142	.3031	.4277	1.1955	.7064	.5504
8.8363	.0138	.2714	.4348	1.2481	.6935	.5174
9.8692	.0133	.2350	.4478	1.2997	.6808	.4770
14.6392	.0127	.1532	.5034	1.3553	.6672	.3728
19.4119	.0129	.1102	.5556	1.3198	.6759	.3060
24.7155	.0137	.0829	.6029	1.2521	.6925	.2551
29.5464	.0149	.0674	.6376	1.1853	.7087	.2216
34.1935	.0164	.0575	.6660	1.1234	.7241	.1967
38.9558	.0184	.0504	.6909	1.0624	.7391	.1764
43.0931	.0205	.0460	.7091	1.0148	.7508	.1619
47.3244	.0230	.0427	.7245	.9736	.7609	.1494
51.9440	.0256	.0405	.7353	.9446	.7680	.1398
56.6395	.0278	.0390	.7433	.9237	.7732	.1317
58.4643	.0303	.0377	.7492	.9094	.7767	.1240
62.0009	.0326	.0369	.7523	.9036	.7781	.1177
65.9248	.0350	.0362	.7534	.9053	.7777	.1114
69.7168	.0369	.0358	.7521	.9149	.7753	.1059
73.7379	.0385	.0354	.7486	.9326	.7710	.1006
78.4123	.0398	.0352	.7422	.9613	.7639	.0951
83.0363	.0405	.0350	.7342	.9961	.7554	.0903
87.9977	.0408	.0348	.7243	1.0379	.7451	.0855
93.7281	.0406	.0347	.7121	1.0885	.7327	.0807
99.3535	.0400	.0346	.7004	1.1374	.7207	.0764
105.7126	.0392	.0345	.6882	1.1885	.7081	.0721
111.7056	.0384	.0344	.6783	1.2305	.6978	.0685
117.7826	.0377	.0343	.6701	1.2658	.6892	.0652
124.5287	.0370	.0342	.6633	1.2961	.6817	.0618
130.9771	.0365	.0342	.6588	1.3159	.6766	.0589
138.2860	.0361	.0341	.6558	1.3324	.6728	.0560
145.4138	.0358	.0340	.6544	1.3409	.6707	.0534
152.9984	.0356	.0339	.6542	1.3448	.6698	.0508
161.8537	.0354	.0338	.6550	1.3451	.6697	.0482
170.7570	.0354	.0338	.6563	1.3429	.6702	.0458
181.3885	.0354	.0337	.6581	1.3389	.6712	.0432
200.3790	.0354	.0336	.6612	1.3319	.6729	.0392

MACH NO = 3.50 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8638	.0142	.0919	1.1617	-.0703	1.0198	1.0098
1.0226	.0152	.0958	.9868	.0066	.9991	.9872
1.2177	.0164	.0155	.8578	.0855	.9760	.9611
1.5378	.0180	.8559	.7283	.1924	.9459	.9213
1.9363	.0197	.7909	.6464	.2999	.9157	.8761
2.4234	.0211	.7220	.5971	.4075	.8854	.8265
3.0076	.0222	.6545	.5587	.5137	.8556	.7740
3.6965	.0232	.5893	.5409	.6110	.8283	.7200
4.4968	.0240	.5265	.5341	.6977	.8039	.6661
5.4147	.0247	.4705	.5340	.7738	.7825	.6134
6.4564	.0254	.4209	.5393	.8371	.7647	.5628
7.6280	.0260	.3751	.5481	.8879	.7504	.5151
8.9362	.0266	.3417	.5584	.9283	.7391	.4706
10.3877	.0271	.3112	.5697	.9605	.7300	.4293
11.9891	.0277	.2859	.5798	.9861	.7228	.3915
13.7486	.0282	.2660	.5897	1.0069	.7170	.3569
15.6766	.0286	.2480	.5987	1.0239	.7122	.3255
17.7798	.0290	.2341	.6068	1.0379	.7083	.2969
20.0690	.0294	.2227	.6140	1.0496	.7050	.2710
22.5545	.0297	.2135	.6205	1.0595	.7022	.2476
25.2479	.0300	.2060	.6263	1.0680	.6998	.2264
28.1612	.0303	.1999	.6314	1.0753	.6977	.2072
31.3076	.0306	.1949	.6359	1.0816	.6960	.1898
34.7011	.0308	.1909	.6400	1.0872	.6944	.1740
38.3570	.0310	.1876	.6436	1.0920	.6931	.1597
42.2915	.0312	.1848	.6468	1.0962	.6919	.1468
46.5227	.0314	.1826	.6497	1.1000	.6908	.1350
51.0693	.0315	.1808	.6523	1.1033	.6899	.1243
55.9504	.0317	.1793	.6546	1.1064	.6890	.1145
61.1910	.0318	.1781	.6567	1.1092	.6882	.1056
66.8151	.0319	.1770	.6586	1.1118	.6875	.0975
72.8501	.0320	.1762	.6602	1.1142	.6868	.0900
79.3261	.0321	.1755	.6617	1.1165	.6862	.0832
86.2768	.0322	.1749	.6630	1.1187	.6856	.0770
93.7393	.0322	.1744	.6642	1.1207	.6850	.0712
101.7546	.0323	.1740	.6652	1.1227	.6844	.0659
110.3675	.0323	.1737	.6662	1.1247	.6839	.0610
119.6272	.0324	.1734	.6670	1.1265	.6834	.0566
129.5867	.0324	.1732	.6678	1.1281	.6829	.0524
140.3035	.0324	.1730	.6685	1.1296	.6825	.0486
151.8395	.0324	.1728	.6692	1.1309	.6821	.0450
164.2613	.0324	.1726	.6698	1.1320	.6818	.0417
177.6398	.0325	.1725	.6705	1.1328	.6816	.0387
192.0520	.0325	.1724	.6711	1.1335	.6814	.0359
207.5886	.0325	.1724	.6716	1.1339	.6813	.0339

MACH NO = 5.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0139	.3451	1.1617	-.0703	1.0198	1.0098
1.6323	.0149	.3075	.9776	.0114	.9968	.9858
1.2502	.0158	.8633	.8331	.0999	.9719	.9569
1.5201	.0168	.8133	.7253	.1928	.9458	.9234
1.9770	.0181	.7392	.6265	.3219	.9095	.8717
2.4008	.0188	.6799	.5770	.4207	.8818	.8288
3.0828	.0195	.5996	.5348	.5505	.8453	.7677
3.5842	.0198	.5415	.5174	.6409	.8198	.7209
4.6044	.0202	.4696	.5083	.7465	.7902	.6595
5.3863	.0204	.4210	.5099	.8116	.7719	.6149
6.5644	.0208	.3638	.5198	.8790	.7529	.5585
7.5272	.0211	.3268	.5311	.9158	.7426	.5189
8.0577	.0217	.2845	.5488	.9495	.7331	.4699
10.5482	.0224	.2497	.5676	.9698	.7274	.4252
12.7669	.0235	.2153	.5899	.9827	.7238	.3754
14.7141	.0244	.1939	.6056	.9879	.7223	.3405
16.9145	.0253	.1768	.6187	.9917	.7213	.3094
19.6678	.0264	.1604	.6317	.9968	.7198	.2753
22.1035	.0272	.1503	.6395	1.0021	.7183	.2515
24.7217	.0279	.1423	.6456	1.0086	.7165	.2302
27.2193	.0288	.1345	.6512	1.0179	.7139	.2068
31.2050	.0293	.1296	.6545	1.0258	.7117	.1903
34.3660	.0298	.1257	.6571	1.0339	.7094	.1755
38.5732	.0304	.1219	.6596	1.0437	.7066	.1590
42.1437	.0307	.1194	.6612	1.0512	.7045	.1472
45.9118	.0311	.1174	.6625	1.0583	.7025	.1366
50.9091	.0314	.1154	.6638	1.0665	.7002	.1246
55.1499	.0317	.1142	.6647	1.0726	.6985	.1160
59.6215	.0319	.1131	.6655	1.0783	.6969	.1081
65.5621	.0321	.1120	.6663	1.0849	.6951	.0992
70.6206	.0323	.1113	.6668	1.0898	.6937	.0926
75.9759	.0324	.1107	.6673	1.0945	.6923	.0866
83.1313	.0325	.1101	.6678	1.1001	.6908	.0797
89.2560	.0326	.1097	.6681	1.1044	.6896	.0746
95.7686	.0327	.1094	.6683	1.1084	.6885	.0698
104.5001	.0327	.1090	.6686	1.1131	.6871	.0643
111.9952	.0328	.1088	.6688	1.1165	.6862	.0602
119.9794	.0328	.1086	.6691	1.1196	.6853	.0564
130.6951	.0328	.1084	.6694	1.1229	.6844	.0520
139.9024	.0328	.1082	.6697	1.1252	.6837	.0487
149.7159	.0329	.1081	.6701	1.1272	.6832	.0456
162.8946	.0329	.1080	.6705	1.1296	.6826	.0421
174.2234	.0329	.1079	.6709	1.1306	.6822	.0394
186.3017	.0329	.1078	.6713	1.1318	.6819	.0370
202.5295	.0329	.1077	.6718	1.1331	.6815	.0341

MACH NO = 10.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.00

L/RN	CA	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	YVCP/LV	RN/RB
.8608	.0137	.9086	1.1617	-.0703	1.0198	1.0098
1.0242	.0144	.8732	.9838	.0082	.9977	.9869
1.2607	.0153	.8258	.8242	.1059	.9702	.9556
1.6473	.0162	.7566	.6816	.2382	.9330	.9084
2.0025	.0167	.7006	.6091	.3401	.9044	.8690
2.5638	.0170	.6244	.5437	.4757	.8663	.8133
3.2303	.0171	.5496	.5029	.6058	.8294	.7557
4.0056	.0168	.4791	.4793	.7281	.7953	.6982
4.6595	.0166	.4303	.4707	.8030	.7726	.6562
5.6218	.0163	.3718	.4690	.8994	.7472	.6026
6.6860	.0160	.3210	.4761	.9683	.7278	.5528
7.9459	.0159	.2880	.4855	1.0059	.7177	.5182
8.7685	.0158	.2501	.5017	1.0394	.7078	.4758
10.7483	.0160	.2048	.5304	1.0605	.7019	.4202
12.8811	.0164	.1706	.5603	1.0568	.7029	.3732
15.1467	.0172	.1448	.5889	1.0385	.7081	.3336
17.5307	.0182	.1253	.6147	1.0132	.7152	.3000
20.0243	.0193	.1104	.6371	.9855	.7227	.2715
22.6229	.0207	.0990	.6557	.9620	.7296	.2470
25.3249	.0222	.0902	.6706	.9417	.7353	.2258
28.1302	.0237	.0835	.6823	.9267	.7395	.2074
31.0390	.0253	.0782	.6909	.9170	.7422	.1911
34.0687	.0270	.0735	.6977	.9124	.7436	.1746
37.7133	.0284	.0705	.7012	.9132	.7433	.1621
40.9792	.0297	.0681	.7031	.9177	.7420	.1509
44.3875	.0309	.0662	.7038	.9250	.7400	.1407
47.9752	.0320	.0648	.7035	.9345	.7373	.1314
51.7964	.0329	.0636	.7024	.9457	.7342	.1227
55.9181	.0337	.0627	.7007	.9588	.7305	.1146
60.4150	.0344	.0619	.6982	.9740	.7262	.1068
65.3680	.0349	.0613	.6949	.9918	.7212	.0994
70.8673	.0353	.0608	.6906	1.0124	.7154	.0923
76.9183	.0354	.0604	.6858	1.0347	.7092	.0856
83.3920	.0354	.0601	.6810	1.0566	.7030	.0794
90.3287	.0352	.0598	.6766	1.0767	.6974	.0737
97.7748	.0351	.0596	.6730	1.0941	.6925	.0684
105.7757	.0349	.0594	.6702	1.1084	.6884	.0636
114.3808	.0347	.0592	.6683	1.1196	.6853	.0590
123.6336	.0346	.0591	.6671	1.1280	.6829	.0548
133.5838	.0345	.0590	.6665	1.1341	.6812	.0509
144.2835	.0343	.0589	.6663	1.1384	.6800	.0473
155.7892	.0343	.0588	.6664	1.1414	.6792	.0439
168.1622	.0342	.0587	.6668	1.1434	.6786	.0408
181.4679	.0341	.0586	.6673	1.1446	.6783	.0379
200.7827	.0340	.0585	.6682	1.1453	.6781	.0344

MACH NO = 15.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0137	.9016	1.1617	-.0703	1.0198	1.0098
1.0204	.0143	.8671	.9870	.0065	.9982	.9875
1.2518	.0151	.8208	.8282	.1029	.9711	.9567
1.6323	.0160	.7524	.6838	.2349	.9340	.9102
2.1063	.0165	.6789	.5905	.3701	.8960	.8581
2.6771	.0167	.6041	.5300	.5051	.8580	.8029
3.1716	.0165	.5491	.4995	.6036	.8303	.7604
3.9210	.0163	.4796	.4730	.7275	.7955	.7041
4.7721	.0158	.4162	.4596	.8373	.7646	.6494
5.7201	.0154	.3600	.4568	.9285	.7390	.5977
6.7571	.0150	.3114	.4621	.9992	.7191	.5498
7.5878	.0147	.2798	.4700	1.0388	.7080	.5166
8.7577	.0145	.2435	.4842	1.0755	.6977	.4762
11.6024	.0144	.1823	.5238	1.1051	.6894	.4000
14.3137	.0148	.1455	.5607	1.0913	.6933	.3471
17.1389	.0155	.1198	.5949	1.0591	.7023	.3051
20.3987	.0167	.0997	.6279	1.0159	.7144	.2677
23.3301	.0181	.0871	.6520	.9785	.7250	.2411
26.2650	.0196	.0779	.6719	.9449	.7344	.2193
29.1849	.0213	.0711	.6880	.9160	.7425	.2012
32.4402	.0233	.0656	.7022	.8899	.7499	.1842
35.3076	.0252	.0619	.7118	.8725	.7548	.1715
38.1593	.0270	.0592	.7189	.8605	.7581	.1605
41.3712	.0291	.0569	.7245	.8528	.7603	.1496
44.2596	.0308	.0553	.7275	.8507	.7609	.1411
47.2168	.0324	.0541	.7289	.8528	.7603	.1332
50.6800	.0341	.0531	.7287	.8601	.7582	.1251
53.9344	.0353	.0524	.7270	.8712	.7551	.1184
57.4132	.0364	.0518	.7238	.8871	.7506	.1119
61.1759	.0372	.0514	.7189	.9081	.7447	.1056
65.8215	.0377	.0510	.7117	.9377	.7364	.0988
70.3673	.0379	.0508	.7039	.9683	.7279	.0929
75.3156	.0377	.0505	.6955	1.0012	.7186	.0873
81.2225	.0373	.0503	.6863	1.0372	.7095	.0814
86.7564	.0369	.0502	.6790	1.0661	.7003	.0766
92.6452	.0365	.0500	.6728	1.0912	.6933	.0720
99.8171	.0360	.0498	.6675	1.1139	.6869	.0671
106.8089	.0356	.0497	.6642	1.1291	.6826	.0630
114.9322	.0353	.0496	.6623	1.1399	.6796	.0589
123.1677	.0351	.0494	.6615	1.1468	.6777	.0550
134.2528	.0349	.0493	.6616	1.1508	.6765	.0507
145.6303	.0348	.0492	.6625	1.1519	.6762	.0469
158.7908	.0347	.0491	.6637	1.1515	.6763	.0431
175.5174	.0347	.0490	.6653	1.1502	.6767	.0392
200.4304	.0346	.0489	.6672	1.1494	.6772	.0344

MACH NO = 20.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0137	.8990	1.1617	-.0723	1.0198	1.0198
1.0190	.0143	.8548	.9882	.0659	.9983	.9877
1.2484	.0150	.8189	.9298	.1017	.9714	.9572
1.6266	.0153	.7519	.5947	.2335	.9344	.9108
2.0959	.0164	.6779	.5908	.3684	.8964	.8592
2.6626	.0165	.6077	.5236	.5034	.8585	.8144
3.1494	.0164	.5491	.4984	.6021	.8308	.7622
3.8892	.0161	.4799	.4708	.7268	.7967	.7063
4.7290	.0156	.4158	.4562	.8382	.7644	.6520
5.6606	.0151	.3618	.4519	.9318	.7351	.6007
6.6785	.0146	.3124	.4557	1.0053	.7174	.5531
7.4921	.0143	.2808	.4626	1.0473	.7056	.5202
9.6350	.0140	.2445	.4735	1.0873	.6944	.4801
12.0417	.0133	.1727	.5216	1.1244	.6840	.3904
15.6920	.0143	.1290	.5696	1.0982	.6913	.3252
19.1047	.0172	.1037	.6078	1.0541	.7037	.2814
22.8560	.0165	.0856	.6419	1.0027	.7182	.2450
26.2097	.0181	.0746	.6658	.9591	.7304	.2197
29.7620	.0201	.0654	.6888	.9165	.7424	.1979
33.1850	.0224	.0607	.7052	.8809	.7524	.1807
36.1855	.0245	.0570	.7185	.8552	.7590	.1679
39.7901	.0269	.0542	.7287	.8343	.7655	.1561
42.2498	.0291	.0522	.7352	.8217	.7690	.1469
45.3832	.0314	.0507	.7399	.8144	.7711	.1389
48.5600	.0335	.0490	.7421	.8139	.7712	.1300
51.6790	.0353	.0489	.7420	.8194	.7697	.1233
54.9864	.0372	.0483	.7397	.8324	.7660	.1163
58.3350	.0382	.0479	.7354	.8510	.7608	.1103
62.3134	.0391	.0475	.7284	.8790	.7529	.1039
66.6362	.0394	.0473	.7142	.9143	.7430	.0977
70.8743	.0394	.0471	.7093	.9511	.7327	.0923
75.8801	.0389	.0470	.6978	.9941	.7206	.0867
80.6734	.0384	.0468	.6876	1.0322	.7099	.0819
86.1280	.0376	.0467	.6778	1.0696	.6993	.0771
91.8203	.0369	.0466	.6698	1.1007	.6906	.0726
97.2790	.0364	.0464	.6643	1.1229	.6844	.0688
103.6947	.0359	.0463	.6603	1.1406	.6794	.0648
110.0055	.0355	.0462	.6583	1.1510	.6765	.0612
117.6215	.0353	.0460	.6576	1.1572	.6747	.0575
126.1406	.0351	.0459	.6583	1.1590	.6742	.0538
134.8858	.0350	.0458	.6596	1.1579	.6745	.0504
145.9089	.0350	.0457	.6615	1.1551	.6753	.0468
157.5405	.0350	.0456	.6674	1.1521	.6762	.0435
172.5810	.0350	.0455	.6654	1.1492	.6770	.0398
201.9593	.0350	.0454	.6678	1.1468	.6777	.0342

MACH NO = 25.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0137	.8979	1.1617	-.0703	1.0198	1.0098
1.0193	.0143	.8640	.9888	.0057	.9984	.9878
1.2468	.0150	.8182	.8305	.1012	.9716	.9574
1.6237	.0158	.7504	.6852	.2329	.9345	.9112
2.0909	.0163	.6777	.5910	.3676	.8967	.8597
2.6528	.0165	.6076	.5295	.5025	.8588	.8051
3.3134	.0163	.5315	.4896	.6334	.8220	.7491
4.0734	.0159	.4638	.4649	.7559	.7875	.6936
4.9301	.0153	.4025	.4524	.8640	.7571	.6401
5.8772	.0148	.3489	.4497	.9539	.7319	.5899
6.9055	.0144	.3020	.4545	1.0238	.7122	.5435
7.7237	.0141	.2717	.4617	1.0634	.7011	.5116
8.8685	.0138	.2370	.4748	1.1009	.6906	.4727
13.8974	.0135	.1591	.5290	1.1335	.6817	.3729
18.8456	.0141	.1179	.5789	1.0957	.6920	.3089
20.8727	.0152	.0971	.6203	1.0417	.7072	.2654
25.0483	.0169	.0765	.6552	.9838	.7235	.2278
28.7603	.0188	.0668	.6809	.9347	.7373	.2036
32.2714	.0209	.0603	.7013	.8918	.7493	.1850
35.8612	.0235	.0555	.7186	.8538	.7600	.1692
39.0132	.0260	.0525	.7307	.8271	.7675	.1574
42.0563	.0285	.0504	.7395	.8081	.7729	.1475
45.0423	.0310	.0489	.7454	.7966	.7761	.1389
48.2773	.0335	.0478	.7488	.7921	.7774	.1306
51.3288	.0357	.0470	.7495	.7955	.7764	.1237
54.5119	.0375	.0465	.7477	.8063	.7734	.1172
58.1841	.0391	.0461	.7429	.8269	.7676	.1105
61.8280	.0401	.0458	.7358	.8544	.7599	.1046
65.7398	.0405	.0456	.7265	.8894	.7500	.0989
70.2799	.0403	.0455	.7144	.9339	.7375	.0931
74.7691	.0398	.0453	.7022	.9786	.7249	.0879
79.4953	.0391	.0452	.6901	1.0230	.7125	.0830
84.4033	.0382	.0451	.6791	1.0636	.7010	.0785
89.8871	.0373	.0450	.6694	1.1005	.6907	.0741
95.1543	.0366	.0449	.6627	1.1268	.6833	.0702
100.6929	.0361	.0448	.6581	1.1458	.6779	.0666
107.0946	.0356	.0446	.6555	1.1586	.6743	.0629
113.4776	.0353	.0445	.6549	1.1643	.6727	.0595
120.4371	.0352	.0444	.6556	1.1654	.6724	.0562
128.9002	.0351	.0442	.6574	1.1630	.6731	.0527
137.4853	.0351	.0441	.6596	1.1588	.6743	.0495
147.3504	.0351	.0440	.6620	1.1541	.6756	.0463
158.7123	.0352	.0439	.6642	1.1500	.6768	.0432
173.1827	.0352	.0439	.6661	1.1468	.6777	.0397
201.4072	.0352	.0437	.6682	1.1453	.6781	.0343

MACH NO = 30.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0137	.8973	1.1617	-.0703	1.0198	1.0098
1.0180	.0143	.8634	.9891	.0055	.9985	.9878
1.2459	.0150	.8177	.8309	.1009	.9717	.9575
1.6221	.0158	.7501	.6855	.2325	.9346	.9114
2.0881	.0163	.6775	.5911	.3671	.8968	.8600
2.6495	.0164	.6036	.5294	.5020	.8589	.8055
3.3059	.0162	.5315	.4893	.6330	.8221	.7496
4.0643	.0158	.4640	.4643	.7557	.7876	.6942
4.9176	.0153	.4028	.4515	.8643	.7570	.6409
5.8604	.0147	.3488	.4483	.9548	.7316	.5907
6.8834	.0143	.3023	.4527	1.0255	.7117	.5445
7.6970	.0140	.2720	.4597	1.0658	.7004	.5126
8.8344	.0137	.2373	.4723	1.1042	.6896	.4737
13.1502	.0133	.1550	.5300	1.1373	.6803	.3680
17.3886	.0140	.1131	.5826	1.0954	.6921	.3018
21.9454	.0153	.0872	.6278	1.0331	.7096	.2529
26.0108	.0169	.0728	.6601	.9768	.7254	.2210
30.1194	.0191	.0631	.6879	.9212	.7411	.1960
33.6712	.0214	.0573	.7083	.8768	.7536	.1785
37.2464	.0242	.0531	.7253	.8383	.7644	.1638
40.6215	.0270	.0503	.7377	.8098	.7724	.1520
43.6372	.0297	.0485	.7458	.7919	.7774	.1428
46.8274	.0324	.0472	.7513	.7815	.7803	.1342
49.8030	.0348	.0463	.7536	.7798	.7808	.1271
53.1011	.0371	.0456	.7532	.7865	.7789	.1200
56.3254	.0389	.0452	.7501	.8012	.7748	.1138
60.0320	.0403	.0449	.7437	.8267	.7676	.1075
63.7059	.0410	.0447	.7351	.8589	.7586	.1018
67.9465	.0411	.0446	.7234	.9019	.7465	.0960
72.1317	.0407	.0445	.7110	.9470	.7338	.0909
76.9106	.0399	.0444	.6970	.9978	.7195	.0856
81.5643	.0389	.0443	.6845	1.0432	.7058	.0811
86.7453	.0379	.0442	.6729	1.0861	.6947	.0766
91.7019	.0371	.0441	.6645	1.1181	.6857	.0727
97.2531	.0363	.0439	.6581	1.1436	.6786	.0688
102.6779	.0358	.0438	.6546	1.1591	.6742	.0654
108.8973	.0354	.0437	.6531	1.1680	.6717	.0618
115.1128	.0352	.0436	.6534	1.1704	.6710	.0587
122.3998	.0351	.0434	.6550	1.1683	.6716	.0553
129.8519	.0351	.0433	.6573	1.1638	.6729	.0523
138.9057	.0352	.0432	.6601	1.1578	.6746	.0491
148.2065	.0353	.0431	.6626	1.1524	.6761	.0461
159.8244	.0353	.0430	.6649	1.1477	.6774	.0429
172.3836	.0354	.0430	.6666	1.1451	.6781	.0398
200.3518	.0353	.0429	.6684	1.1444	.6793	.0345

MACH NO = 3.50 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	XCP/D	XVCP/LV	RN/RB
.8436	.0141	.3961	1.1854	-.0792	1.0251	1.0125
1.0016	.0151	.3967	1.0077	-.0038	1.0012	.9874
1.1915	.0162	.3932	.8729	.0726	.9770	.9590
1.5019	.0178	.8497	.7461	.1746	.9447	.9158
1.8868	.0194	.7815	.6634	.2754	.9127	.8674
2.3549	.0207	.7114	.6099	.3743	.8814	.8150
2.7653	.0215	.6591	.5827	.4456	.8585	.7740
3.3964	.0225	.5919	.5617	.5347	.8306	.7184
4.1286	.0234	.5295	.5527	.6123	.8061	.6631
4.9667	.0241	.4735	.5513	.6792	.7849	.6095
5.9153	.0248	.4243	.5556	.7339	.7675	.5584
6.7019	.0253	.3920	.5614	.7673	.7569	.5220
7.8545	.0259	.3548	.5708	.8034	.7455	.4766
9.4688	.0267	.3168	.5835	.8378	.7346	.4248
10.9052	.0273	.2924	.5933	.8590	.7279	.3874
12.4764	.0279	.2723	.6025	.8763	.7224	.3533
14.6390	.0285	.2525	.6126	.8937	.7169	.3152
16.5356	.0289	.2399	.6197	.9052	.7133	.2879
18.5878	.0294	.2298	.6260	.9152	.7101	.2633
20.8030	.0297	.2215	.6315	.9239	.7073	.2410
23.8137	.0301	.2134	.6374	.9333	.7044	.2162
26.4262	.0304	.2082	.6415	.9399	.7023	.1984
29.2304	.0307	.2040	.6452	.9457	.7004	.1824
33.0210	.0309	.1998	.6492	.9521	.6984	.1644
36.2956	.0311	.1972	.6520	.9565	.6970	.1515
39.7989	.0313	.1950	.6545	.9605	.6957	.1397
44.5208	.0315	.1928	.6573	.9649	.6943	.1265
48.5915	.0317	.1914	.6593	.9681	.6933	.1170
52.9418	.0318	.1902	.6611	.9711	.6924	.1082
57.5913	.0319	.1893	.6626	.9738	.6915	.1002
63.8585	.0320	.1883	.6644	.9771	.6905	.0912
69.2670	.0321	.1877	.6656	.9795	.6897	.0846
75.0571	.0321	.1872	.6667	.9818	.6890	.0785
82.8791	.0322	.1866	.6679	.9845	.6881	.0715
89.6456	.0322	.1863	.6687	.9865	.6875	.0664
96.9047	.0322	.1860	.6695	.9883	.6870	.0617
106.7317	.0323	.1857	.6705	.9902	.6863	.0563
115.2479	.0323	.1855	.6713	.9914	.6860	.0523
124.3954	.0323	.1853	.6720	.9924	.6856	.0486
134.2231	.0323	.1851	.6727	.9932	.6854	.0452
147.5441	.0323	.1850	.6736	.9939	.6852	.0413
159.0987	.0324	.1849	.6743	.9944	.6850	.0384
171.5169	.0324	.1848	.6750	.9948	.6849	.0357
188.3545	.0324	.1847	.6757	.9952	.6847	.0326
202.9626	.0324	.1847	.6762	.9956	.6846	.0303

MACH NO = 5.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0138	.9492	1.1854	-.0792	1.0251	1.0125
1.0097	.0147	.9084	.9995	.0002	.9999	.9862
1.2206	.0157	.8610	.8533	.0855	.9729	.9547
1.4811	.0166	.8079	.7445	.1738	.9450	.9186
1.9186	.0178	.7304	.6449	.2942	.9068	.8636
2.3208	.0185	.6695	.5953	.3844	.8782	.8186
2.9551	.0191	.5896	.5538	.4986	.8420	.7564
3.5032	.0195	.5330	.5369	.5758	.8176	.7098
4.3339	.0198	.4638	.5279	.6641	.7896	.6491
5.0331	.0201	.4174	.5290	.7178	.7726	.6056
6.0682	.0205	.3632	.5379	.7724	.7553	.5509
6.9225	.0209	.3282	.5481	.8019	.7460	.5127
8.1658	.0215	.2884	.5642	.8286	.7375	.4656
9.8964	.0225	.2485	.5855	.8471	.7317	.4130
11.4262	.0237	.2231	.6016	.8544	.7294	.3754
13.5345	.0245	.1981	.6194	.8592	.7278	.3336
15.8701	.0257	.1791	.6338	.8629	.7267	.2969
18.4008	.0268	.1648	.6446	.8678	.7251	.2654
20.5628	.0276	.1561	.6510	.8729	.7235	.2433
23.4388	.0285	.1477	.6568	.8807	.7210	.2190
26.5109	.0293	.1413	.6609	.8895	.7182	.1979
29.7839	.0300	.1365	.6638	.8986	.7153	.1795
32.5513	.0304	.1335	.6655	.9059	.7130	.1664
36.2040	.0309	.1305	.6671	.9147	.7102	.1518
40.3826	.0313	.1281	.6683	.9230	.7076	.1388
43.3586	.0316	.1267	.6690	.9293	.7056	.1295
47.6881	.0319	.1252	.6698	.9366	.7033	.1189
52.3038	.0322	.1240	.6704	.9434	.7012	.1094
57.2389	.0324	.1230	.6708	.9498	.6991	.1008
61.4415	.0325	.1224	.6711	.9547	.6976	.0945
67.0456	.0326	.1218	.6713	.9605	.6957	.0872
73.0760	.0327	.1213	.6714	.9661	.6940	.0805
79.5727	.0328	.1208	.6715	.9713	.6923	.0743
85.1335	.0328	.1206	.6715	.9752	.6911	.0697
92.5748	.0329	.1203	.6716	.9795	.6897	.0644
100.6043	.0329	.1200	.6718	.9832	.6885	.0596
107.4844	.0329	.1198	.6721	.9858	.6877	.0559
116.6997	.0329	.1197	.6724	.9885	.6869	.0517
126.6519	.0329	.1195	.6728	.9908	.6862	.0478
137.4021	.0329	.1194	.6732	.9926	.6856	.0442
146.6209	.0329	.1193	.6736	.9939	.6852	.0415
158.9768	.0329	.1193	.6740	.9952	.6847	.0384
172.3281	.0329	.1192	.6745	.9963	.6844	.0355
186.7562	.0329	.1191	.6750	.9972	.6841	.0329
202.3486	.0328	.1191	.6754	.9980	.6839	.0304

MACH NO = 10.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RR
.8436	.0136	.9127	1.1854	-.0792	1.0251	1.0125
1.0004	.0143	.8746	1.0071	-.0035	1.0011	.9876
1.2251	.0151	.8245	.8475	.0891	.9718	.9541
1.5915	.0160	.7520	.7038	.2131	.9325	.9040
1.9193	.0165	.6952	.6318	.3051	.9034	.8635
2.4316	.0168	.6192	.5667	.4251	.8654	.8070
3.0338	.0168	.5455	.5259	.5389	.8293	.7493
3.7272	.0167	.4767	.5021	.6423	.7965	.6923
4.3063	.0165	.4294	.4933	.7102	.7750	.6510
5.1536	.0162	.3730	.4910	.7852	.7513	.5987
6.0810	.0161	.3242	.4971	.8414	.7335	.5503
6.8244	.0160	.2925	.5056	.8718	.7238	.5168
7.8729	.0160	.2562	.5205	.8984	.7154	.4760
9.8425	.0164	.2066	.5514	.9150	.7102	.4144
11.9513	.0171	.1708	.5832	.9066	.7128	.3640
14.1702	.0181	.1449	.6127	.8857	.7195	.3227
16.1448	.0192	.1283	.6348	.8642	.7262	.2932
18.5215	.0207	.1138	.6564	.8401	.7339	.2640
20.9695	.0224	.1031	.6736	.8197	.7403	.2395
23.4802	.0241	.0951	.6866	.8045	.7452	.2187
26.0538	.0259	.0892	.6961	.7948	.7482	.2008
28.6905	.0276	.0847	.7026	.7902	.7497	.1852
31.0064	.0289	.0817	.7063	.7899	.7498	.1735
33.7883	.0304	.0791	.7087	.7931	.7488	.1611
36.6848	.0317	.0771	.7096	.7993	.7468	.1501
39.7395	.0329	.0755	.7092	.8083	.7440	.1399
43.0080	.0339	.0743	.7077	.8198	.7403	.1304
46.0283	.0346	.0735	.7056	.8320	.7364	.1228
49.8709	.0353	.0727	.7018	.8494	.7309	.1142
54.1349	.0357	.0721	.6968	.8703	.7243	.1061
58.9115	.0358	.0717	.6908	.8942	.7167	.0982
64.7080	.0357	.0713	.6844	.9193	.7088	.0906
70.3490	.0354	.0709	.6784	.9430	.7013	.0834
76.0179	.0352	.0707	.6742	.9604	.6958	.0776
83.1627	.0349	.0704	.6707	.9763	.6908	.0713
90.9198	.0347	.0702	.6686	.9877	.6871	.0656
99.3417	.0345	.0700	.6675	.9955	.6847	.0603
108.4845	.0344	.0698	.6673	1.0006	.6830	.0554
116.9412	.0342	.0697	.6675	1.0034	.6822	.0516
127.5894	.0341	.0696	.6681	1.0052	.6816	.0475
139.1483	.0341	.0695	.6690	1.0060	.6813	.0437
151.6964	.0340	.0694	.6700	1.0061	.6813	.0402
165.3190	.0340	.0693	.6710	1.0059	.6814	.0370
180.1095	.0339	.0693	.6719	1.0055	.6815	.0340
201.0065	.0339	.0692	.6731	1.0049	.6817	.0306

MACH NO = 15.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8438	.0136	.0057	1.1854	-.0792	1.0251	1.0125
.9967	.0142	.0086	1.0103	-.0051	1.0016	.9882
1.2156	.0150	.0197	.8521	.0859	.9728	.9555
1.5725	.0158	.0488	.7075	.2095	.9340	.9065
2.0077	.0163	.0745	.6144	.3302	.8954	.8532
2.5264	.0164	.0900	.5539	.4493	.8577	.7973
3.0719	.0164	.0958	.5233	.5347	.8306	.7548
3.6411	.0161	.0778	.4967	.6404	.7971	.6989
4.3942	.0158	.0662	.4831	.7326	.7679	.6451
5.2253	.0154	.0620	.4799	.8080	.7440	.5946
6.1267	.0151	.0752	.4845	.8656	.7258	.5481
6.8432	.0149	.0848	.4918	.8973	.7157	.5160
7.8449	.0148	.0899	.5049	.9262	.7066	.4770
10.5253	.0150	.1057	.5462	.9473	.6999	.3966
13.0772	.0156	.1482	.5845	.9296	.7058	.3418
15.6963	.0167	.1228	.6188	.8957	.7163	.2994
18.6252	.0183	.1038	.6501	.8566	.7296	.2629
21.2427	.0200	.0921	.6724	.8249	.7387	.2370
23.8182	.0219	.0837	.6901	.7977	.7473	.2161
26.6195	.0241	.0772	.7054	.7732	.7551	.1972
29.0902	.0261	.0730	.7159	.7567	.7603	.1831
31.5284	.0282	.0700	.7235	.7454	.7639	.1710
34.2282	.0304	.0675	.7292	.7385	.7661	.1594
36.6834	.0322	.0659	.7321	.7373	.7664	.1501
39.2003	.0339	.0647	.7331	.7407	.7654	.1416
42.1194	.0356	.0637	.7319	.7503	.7623	.1329
44.9078	.0367	.0630	.7288	.7643	.7579	.1255
47.9037	.0376	.0625	.7238	.7838	.7517	.1185
51.5349	.0381	.0621	.7160	.8117	.7429	.1109
55.1131	.0382	.0619	.7074	.8415	.7334	.1043
58.9986	.0380	.0616	.6979	.8739	.7232	.0981
63.6485	.0375	.0614	.6874	.9098	.7118	.0914
68.0846	.0369	.0612	.6790	.9389	.7026	.0859
72.8055	.0363	.0610	.6722	.9634	.6948	.0807
78.4873	.0358	.0608	.6667	.9847	.6881	.0753
84.1060	.0354	.0606	.6635	.9982	.6838	.0705
90.3358	.0351	.0605	.6620	1.0069	.6810	.0660
98.1711	.0348	.0603	.6619	1.0118	.6795	.0610
106.2662	.0347	.0601	.6628	1.0130	.6791	.0565
115.6083	.0346	.0599	.6644	1.0122	.6794	.0522
127.8216	.0346	.0598	.6665	1.0102	.6800	.0474
140.9041	.0346	.0597	.6684	1.0084	.6806	.0432
155.2774	.0345	.0596	.6699	1.0070	.6810	.0393
173.0121	.0345	.0595	.6714	1.0061	.6813	.0354
201.1299	.0344	.0595	.6729	1.0055	.6815	.0306

MACH NO = 20.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	YYCP/LV	RN/RB
.8436	.0136	.9070	1.1954	-.0792	1.0251	1.0125
.9954	.0142	.8663	1.0115	-.0057	1.0018	.9984
1.2121	.0149	.8179	.8539	.0847	.9732	.9560
1.5656	.0157	.7476	.7089	.2068	.9345	.9074
1.9963	.0162	.6738	.6151	.3283	.8960	.8545
2.5091	.0163	.5998	.5539	.4472	.8583	.7991
2.9491	.0162	.5460	.5225	.5329	.8312	.7569
3.6095	.0159	.4793	.4949	.6394	.7975	.7014
4.3515	.0155	.4170	.4801	.7330	.7678	.6480
5.1690	.0151	.3629	.4755	.8104	.7433	.5978
6.0538	.0148	.3162	.4788	.8702	.7243	.5516
6.7567	.0146	.2858	.4851	.9039	.7137	.5197
7.7748	.0144	.2509	.4971	.9354	.7037	.4810
10.8813	.0144	.1767	.5448	.9609	.6956	.3880
13.9123	.0152	.1360	.5896	.9338	.7042	.3271
17.2445	.0166	.1036	.6317	.8880	.7187	.2789
20.2394	.0182	.0927	.6602	.8469	.7317	.2463
23.4797	.0203	.0813	.6854	.8069	.7444	.2194
26.1239	.0225	.0743	.7037	.7754	.7544	.2003
28.9804	.0251	.0692	.7192	.7475	.7632	.1837
31.4987	.0275	.0659	.7298	.7284	.7693	.1711
34.1771	.0301	.0634	.7379	.7147	.7736	.1596
36.6106	.0323	.0618	.7425	.7086	.7755	.1503
39.2917	.0346	.0605	.7446	.7090	.7754	.1413
41.8222	.0364	.0597	.7440	.7159	.7732	.1337
44.7237	.0380	.0591	.7416	.7309	.7685	.1260
47.5654	.0391	.0588	.7349	.7517	.7619	.1192
50.9100	.0397	.0585	.7260	.7820	.7523	.1121
54.2185	.0397	.0583	.7159	.8157	.7416	.1059
58.0940	.0393	.0581	.7035	.8566	.7287	.0994
61.8844	.0386	.0580	.5919	.8948	.7166	.0938
66.2423	.0377	.0578	.6802	.9335	.7043	.0881
70.4225	.0369	.0577	.6714	.9635	.6948	.0833
74.8232	.0363	.0575	.6648	.9871	.6873	.0787
79.9660	.0357	.0573	.6601	1.0052	.6816	.0740
85.0828	.0353	.0571	.6580	1.0153	.6784	.0698
91.2301	.0350	.0570	.6577	1.0201	.6768	.0653
97.5247	.0349	.0568	.6589	1.0204	.6768	.0613
105.3133	.0348	.0566	.6612	1.0176	.6777	.0570
113.5278	.0348	.0565	.6636	1.0138	.6789	.0531
123.9901	.0349	.0563	.6663	1.0096	.6802	.0488
135.3268	.0349	.0562	.6684	1.0068	.6811	.0449
150.1272	.0349	.0561	.6701	1.0053	.6815	.0406
166.5243	.0348	.0560	.6713	1.0052	.6816	.0367
201.6065	.0347	.0560	.6728	1.0060	.6813	.0305

MACH NO = 25.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

L/ON	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RO
.8436	.0136	.9020	1.1854	-.0792	1.0251	1.0125
.9948	.0142	.8654	1.0121	-.0059	1.0019	.9885
1.2105	.0149	.8173	.8547	.0841	.9734	.9562
1.5624	.0156	.7472	.7095	.2060	.9347	.9078
1.9911	.0161	.6736	.6154	.3274	.8963	.8551
2.5012	.0162	.5999	.5538	.4463	.8596	.7999
3.0950	.0161	.5288	.5138	.5597	.8227	.7439
3.7718	.0157	.4627	.4891	.6638	.7897	.6890
4.5279	.0153	.4032	.4765	.7544	.7610	.6364
5.3563	.0149	.3510	.4734	.8294	.7376	.5874
6.2487	.0145	.3062	.4777	.8850	.7196	.5424
6.9530	.0143	.2770	.4844	.9166	.7096	.5114
7.9321	.0141	.2436	.4967	.9458	.7004	.4738
11.3230	.0142	.1682	.5480	.9665	.6938	.3777
14.8514	.0151	.1256	.5987	.9293	.7056	.3113
19.7456	.0166	.1005	.6394	.8798	.7213	.2660
21.6944	.0196	.0852	.6703	.8335	.7360	.2331
24.8438	.0208	.0756	.6947	.7919	.7491	.2088
27.7017	.0224	.0693	.7139	.7565	.7604	.1903
30.6691	.0261	.0651	.7286	.7284	.7693	.1756
33.2222	.0289	.0622	.7392	.7084	.7756	.1635
35.8025	.0315	.0603	.7462	.6963	.7794	.1533
38.3635	.0341	.0589	.7500	.6920	.7808	.1443
40.9619	.0363	.0580	.7507	.6954	.7797	.1362
43.6589	.0382	.0574	.7484	.7068	.7761	.1287
46.5159	.0396	.0571	.7432	.7265	.7699	.1216
49.5767	.0404	.0568	.7350	.7544	.7610	.1149
52.8581	.0406	.0566	.7242	.7900	.7497	.1084
56.3563	.0403	.0565	.7115	.8312	.7367	.1022
60.0876	.0396	.0564	.6979	.8751	.7228	.0964
64.0229	.0385	.0563	.6848	.9178	.7093	.0910
68.1340	.0375	.0561	.6734	.9553	.6974	.0859
72.4232	.0366	.0560	.6647	.9853	.6879	.0811
76.9485	.0359	.0558	.6587	1.0058	.6811	.0767
81.7860	.0354	.0556	.6556	1.0200	.6769	.0724
87.0235	.0351	.0555	.6548	1.0260	.6750	.0683
92.7666	.0349	.0553	.6560	1.0264	.6749	.0643
99.1479	.0349	.0551	.6583	1.0230	.6760	.0604
106.3773	.0349	.0549	.6613	1.0176	.6777	.0565
114.5552	.0350	.0548	.6643	1.0120	.6794	.0526
124.0874	.0350	.0547	.6670	1.0074	.6809	.0488
135.3051	.0351	.0546	.6691	1.0047	.6818	.0449
148.6949	.0351	.0545	.6705	1.0038	.6820	.0410
164.9011	.0350	.0544	.6714	1.0045	.6818	.0371
201.2566	.0348	.0543	.6726	1.0066	.6812	.0306

MACH NO = 20.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	XCP/D	XVCP/LV	RN/RB
.8436	.0136	.0013	1.1854	-.0792	1.0251	1.0125
.9944	.0142	.0049	1.0124	-.0061	1.0019	.9886
1.2096	.0140	.0168	.8551	.0838	.9735	.9563
1.5606	.0156	.7469	.7099	.2056	.9349	.9081
1.9881	.0161	.6735	.6156	.3269	.8965	.8555
2.4967	.0162	.5998	.5538	.4458	.8598	.8003
3.0886	.0160	.5280	.5175	.5593	.8228	.7445
3.7639	.0157	.4629	.4886	.6636	.7898	.6896
4.5161	.0153	.4035	.4756	.7545	.7610	.6372
5.3408	.0148	.3513	.4722	.8291	.7374	.5882
6.2287	.0144	.3065	.4761	.8864	.7192	.5473
6.9291	.0142	.2773	.4826	.9185	.7090	.5124
7.9020	.0140	.2430	.4946	.9484	.6996	.4749
11.5328	.0140	.1642	.5402	.9696	.6929	.3330
15.5597	.0151	.1187	.6059	.9240	.7073	.3013
19.2296	.0169	.0950	.6465	.8717	.7239	.2564
22.6884	.0189	.0829	.6773	.8232	.7392	.2248
25.8960	.0213	.0771	.7016	.7797	.7530	.2018
28.8683	.0241	.0764	.7206	.7432	.7646	.1843
31.6525	.0269	.0627	.7349	.7151	.7735	.1704
34.3061	.0299	.0632	.7460	.6958	.7796	.1590
36.8890	.0327	.0585	.7512	.6851	.7830	.1403
39.4592	.0353	.0574	.7541	.6829	.7837	.1408
42.0799	.0375	.0567	.7538	.6889	.7818	.1330
44.8208	.0393	.0562	.7503	.7036	.7771	.1257
47.7354	.0406	.0560	.7436	.7272	.7696	.1188
50.8416	.0412	.0558	.7339	.7595	.7594	.1123
54.1595	.0411	.0557	.7216	.7993	.7468	.1060
57.6844	.0404	.0556	.7076	.8442	.7326	.1001
61.4075	.0394	.0555	.6931	.8905	.7179	.0945
65.3219	.0383	.0554	.6797	.9342	.7041	.0893
69.6928	.0372	.0552	.6678	.9734	.6916	.0841
73.9563	.0363	.0551	.6598	1.0010	.6829	.0796
78.4601	.0356	.0549	.6550	1.0191	.6772	.0753
83.2768	.0352	.0547	.6530	1.0287	.6741	.0712
88.4881	.0349	.0545	.6535	1.0311	.6734	.0673
94.1912	.0348	.0543	.6556	1.0284	.6742	.0634
100.5058	.0349	.0542	.6587	1.0227	.6760	.0596
107.5839	.0350	.0540	.6620	1.0160	.6782	.0559
115.6427	.0351	.0539	.6652	1.0098	.6801	.0522
124.9311	.0352	.0538	.6678	1.0054	.6815	.0484
135.8071	.0352	.0537	.6696	1.0031	.6823	.0447
148.7427	.0352	.0536	.6708	1.0029	.6823	.0410
164.3788	.0351	.0535	.6715	1.0042	.6819	.0372
200.1267	.0349	.0534	.6724	1.0069	.6810	.0307

MACH NO = 3.50 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0140	1.0008	1.2101	-.0882	1.0311	1.0154
.9807	.0150	.9583	1.0293	-.0142	1.0050	.9881
1.1655	.0161	.9119	.8927	.0599	.9789	.9573
1.4666	.0176	.8450	.7644	.1574	.9445	.9110
1.7383	.0187	.7921	.6985	.2288	.9193	.8729
2.1681	.0201	.7204	.6385	.3208	.8860	.8187
2.6812	.0212	.6496	.6001	.4086	.8559	.7623
3.2829	.0221	.5826	.5795	.4868	.8283	.7052
3.7950	.0228	.5361	.5717	.5389	.8100	.6630
4.5617	.0235	.4798	.5684	.5990	.7887	.6085
5.4274	.0242	.4308	.5714	.6475	.7717	.5568
6.1433	.0248	.3988	.5764	.6768	.7613	.5202
7.1892	.0255	.3622	.5850	.7081	.7503	.4747
8.6477	.0263	.3250	.5968	.7376	.7399	.4230
9.9394	.0270	.3012	.6059	.7557	.7335	.3858
11.7158	.0277	.2775	.6164	.7736	.7272	.3442
13.6791	.0284	.2594	.6254	.7881	.7221	.3076
15.3890	.0289	.2480	.6316	.7979	.7186	.2815
17.7114	.0294	.2368	.6383	.8085	.7149	.2524
19.7213	.0298	.2297	.6428	.8161	.7122	.2317
22.4340	.0302	.2228	.6476	.8245	.7092	.2096
25.3810	.0306	.2174	.6515	.8319	.7066	.1882
27.9166	.0308	.2140	.6545	.8371	.7048	.1736
31.3222	.0311	.2106	.6576	.8429	.7028	.1572
34.2467	.0313	.2085	.6598	.8470	.7013	.1454
38.1702	.0315	.2063	.6622	.8517	.6997	.1321
42.4142	.0316	.2046	.6643	.8559	.6982	.1202
46.0595	.0317	.2035	.6658	.8590	.6971	.1116
50.9570	.0319	.2024	.6674	.8626	.6958	.1018
56.2590	.0319	.2015	.6688	.8660	.6946	.0929
60.8458	.0320	.2009	.6698	.8685	.6937	.0864
67.0152	.0321	.2004	.6709	.8714	.6927	.0790
72.7410	.0321	.2000	.6717	.8735	.6920	.0736
79.5316	.0321	.1996	.6727	.8756	.6912	.0673
87.3686	.0321	.1992	.6737	.8774	.6905	.0616
94.1450	.0322	.1990	.6745	.8785	.6902	.0573
103.3047	.0322	.1988	.6755	.8796	.6898	.0525
111.2284	.0322	.1987	.6763	.8803	.6895	.0489
121.9422	.0322	.1985	.6771	.8811	.6893	.0448
133.6330	.0323	.1984	.6780	.8818	.6890	.0410
143.7496	.0323	.1983	.6785	.8823	.6888	.0382
157.4325	.0323	.1982	.6792	.8829	.6886	.0350
169.2747	.0323	.1982	.6797	.8834	.6885	.0326
185.2935	.0323	.1981	.6803	.8839	.6883	.0298
202.7796	.0323	.1981	.6808	.8844	.6881	.0273

MACH NO = 5.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0137	.9538	1.2101	-.0882	1.0311	1.0154
.9872	.0146	.9102	1.0223	-.0109	1.0038	.9870
1.1184	.0152	.8772	.9190	.0437	.9846	.9650
1.4424	.0165	.8040	.7643	.1554	.9452	.9145
1.7472	.0173	.7442	.6848	.2400	.9154	.8717
2.2373	.0182	.6628	.6150	.3491	.8769	.8106
2.5640	.0187	.6037	.5820	.4254	.8500	.7640
3.0149	.0192	.5297	.5569	.5159	.8181	.7024
3.8657	.0195	.4791	.5485	.5739	.7976	.6576
4.6844	.0199	.4188	.5476	.6363	.7756	.6006
5.3612	.0202	.3793	.5527	.6720	.7630	.5604
6.0916	.0206	.3444	.5608	.6991	.7535	.5227
7.1480	.0212	.3048	.5747	.7240	.7447	.4763
8.6018	.0221	.2649	.5938	.7415	.7385	.4245
10.2063	.0232	.2339	.6123	.7498	.7356	.3790
11.9681	.0243	.2100	.6285	.7537	.7342	.3391
14.3050	.0258	.1886	.6441	.7572	.7330	.2975
16.4552	.0269	.1752	.6539	.7613	.7315	.2673
18.7968	.0279	.1651	.6609	.7672	.7294	.2408
21.2907	.0289	.1575	.6658	.7745	.7269	.2177
23.9336	.0296	.1518	.6691	.7827	.7240	.1977
26.7292	.0303	.1474	.6713	.7912	.7210	.1801
29.6833	.0308	.1441	.6728	.7997	.7180	.1647
33.4505	.0314	.1411	.6739	.8094	.7146	.1484
36.7916	.0318	.1392	.6746	.8171	.7118	.1365
40.3372	.0321	.1377	.6750	.8244	.7093	.1258
44.1120	.0323	.1366	.6752	.8313	.7068	.1160
48.1436	.0325	.1356	.6753	.8379	.7045	.1072
52.4608	.0327	.1349	.6753	.8443	.7023	.0991
58.0586	.0328	.1342	.6751	.8516	.6997	.0903
63.1051	.0329	.1337	.6748	.8574	.6976	.0836
68.5306	.0329	.1333	.6747	.8626	.6953	.0774
74.3673	.0329	.1330	.6746	.8673	.6942	.0717
80.6496	.0329	.1327	.6746	.8712	.6928	.0664
87.4143	.0329	.1325	.6748	.8745	.6916	.0615
94.7011	.0329	.1323	.6751	.8772	.6907	.0570
104.1941	.0329	.1321	.6756	.8798	.6897	.0521
112.7828	.0328	.1320	.6760	.8815	.6891	.0482
122.0403	.0328	.1319	.6765	.8829	.6886	.0447
132.0196	.0328	.1318	.6770	.8840	.6882	.0415
142.7778	.0328	.1318	.6775	.8850	.6879	.0384
154.3764	.0328	.1317	.6780	.8857	.6877	.0356
166.8817	.0329	.1317	.6786	.8862	.6875	.0330
183.1857	.0328	.1316	.6792	.8866	.6873	.0302
201.0325	.0327	.1316	.6798	.8869	.6872	.0276

MACH NO = 10.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RR
.8264	.0136	.0172	1.2181	-.0882	1.0311	1.0154
.9768	.0142	.0168	1.0314	-.0152	1.0053	.9888
1.1872	.0150	.0251	.8734	.0717	.9747	.9538
1.5261	.0158	.0514	.7391	.1858	.9345	.9024
1.8263	.0162	.0646	.6577	.2692	.9051	.8612
2.2911	.0166	.0192	.5920	.3760	.8674	.8045
2.8324	.0166	.0468	.5502	.4759	.8322	.7471
3.4502	.0165	.0737	.5256	.5653	.8006	.6909
3.9621	.0164	.0338	.5160	.6235	.7801	.6503
4.7059	.0163	.0701	.5126	.6872	.7577	.5992
5.5133	.0162	.0329	.5173	.7347	.7409	.5521
6.1561	.0162	.0013	.5247	.7602	.7319	.5196
7.0564	.0162	.0661	.5380	.7825	.7241	.4800
8.0772	.0162	.0122	.5703	.7964	.7192	.4129
10.7566	.0176	.1798	.5994	.7875	.7223	.3655
12.8697	.0189	.1516	.6297	.7666	.7297	.3217
15.0446	.0206	.1324	.6550	.7431	.7379	.2864
16.9857	.0221	.1203	.6728	.7248	.7444	.2608
19.2380	.0241	.1102	.6884	.7084	.7502	.2363
21.2339	.0257	.1037	.6984	.6986	.7535	.2182
23.5442	.0276	.0981	.7065	.6923	.7559	.2004
25.8952	.0294	.0941	.7117	.6906	.7564	.1850
28.0020	.0308	.0915	.7143	.6926	.7558	.1731
30.4963	.0323	.0893	.7154	.6980	.7538	.1609
32.7875	.0334	.0878	.7151	.7055	.7512	.1510
35.5772	.0345	.0865	.7132	.7174	.7470	.1406
38.6076	.0354	.0855	.7096	.7331	.7415	.1308
41.5119	.0358	.0849	.7051	.7503	.7354	.1226
45.1883	.0361	.0843	.6984	.7737	.7271	.1135
49.3325	.0360	.0838	.6908	.7995	.7180	.1048
53.3915	.0357	.0834	.6843	.8220	.7101	.0975
58.5057	.0354	.0831	.6779	.8448	.7021	.0896
63.4711	.0351	.0828	.6737	.8608	.6964	.0831
69.8511	.0347	.0825	.6705	.8746	.6916	.0760
77.1254	.0345	.0822	.6690	.8840	.6883	.0693
84.1059	.0343	.0820	.6688	.8890	.6865	.0638
92.7973	.0341	.0818	.6693	.8920	.6854	.0581
101.0814	.0340	.0817	.6702	.8931	.6851	.0536
111.3926	.0340	.0815	.6715	.8932	.6850	.0488
122.6900	.0339	.0814	.6729	.8929	.6851	.0445
133.4585	.0339	.0813	.6741	.8924	.6853	.0410
146.8692	.0339	.0813	.6753	.8918	.6855	.0374
159.6557	.0338	.0812	.6763	.8913	.6857	.0345
175.5838	.0338	.0812	.6773	.8908	.6859	.0314
200.0188	.0338	.0811	.6786	.8902	.6861	.0277

MACH NO = 15.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/C	XVCP/LV	RN/RB
.8264	.0135	.9102	1.2101	-.0882	1.0311	1.0154
.9732	.0141	.8708	1.0346	-.0167	1.0059	.9894
1.1786	.0148	.8203	.8779	.0687	.9759	.9552
1.5098	.0156	.7482	.7338	.1817	.9359	.9049
1.9070	.0161	.6738	.6404	.2917	.8971	.8508
2.3766	.0162	.6001	.5793	.3974	.8599	.7948
2.7767	.0162	.5460	.5481	.4722	.8335	.7526
3.3737	.0160	.4806	.5209	.5535	.8013	.6974
4.0788	.0157	.4217	.5066	.6423	.7735	.6446
4.7674	.0155	.3682	.5025	.7060	.7510	.5953
5.5515	.0153	.3228	.5060	.7542	.7340	.5501
6.1706	.0152	.2934	.5124	.7806	.7247	.5189
7.0307	.0151	.2595	.5244	.8043	.7163	.4810
9.9381	.0155	.1922	.5666	.8199	.7109	.3967
11.9153	.0165	.1578	.6059	.7987	.7183	.3401
14.3162	.0180	.1286	.6389	.7666	.7297	.2973
16.6948	.0198	.1119	.6668	.7352	.7407	.2644
19.0208	.0218	.1005	.6878	.7082	.7503	.2385
21.9019	.0241	.0919	.7055	.6839	.7588	.2160
23.6901	.0264	.0866	.7176	.6669	.7648	.1993
25.8314	.0286	.0828	.7265	.6550	.7690	.1854
27.9511	.0308	.0802	.7325	.6482	.7714	.1734
30.0805	.0328	.0782	.7360	.6465	.7720	.1628
32.4761	.0348	.0768	.7369	.6507	.7705	.1523
34.7444	.0363	.0758	.7352	.6603	.7671	.1436
37.1440	.0374	.0752	.7312	.6757	.7617	.1353
39.7236	.0382	.0747	.7248	.6970	.7542	.1275
42.5242	.0384	.0744	.7163	.7236	.7448	.1199
45.8875	.0383	.0741	.7051	.7576	.7328	.1120
49.2107	.0378	.0739	.6944	.7901	.7214	.1051
52.7727	.0371	.0736	.6843	.8211	.7104	.0986
56.5615	.0364	.0734	.6758	.8479	.7010	.0925
61.0546	.0358	.0732	.6689	.8711	.6928	.0862
65.5270	.0353	.0729	.6649	.8859	.6876	.0807
70.4772	.0349	.0727	.6631	.8952	.6843	.0754
75.0374	.0347	.0724	.6609	.8996	.6827	.0702
82.3732	.0345	.0722	.6641	.9004	.6825	.0651
90.4854	.0345	.0720	.6664	.8998	.6830	.0595
99.1776	.0345	.0718	.6688	.8963	.6839	.0546
109.4526	.0344	.0717	.6710	.8940	.6847	.0497
121.6452	.0344	.0716	.6730	.8924	.6853	.0449
135.3651	.0344	.0715	.6745	.8918	.6855	.0405
152.1595	.0343	.0714	.6757	.8916	.6856	.0361
169.1519	.0343	.0713	.6769	.8915	.6856	.0326
200.2234	.0342	.0713	.6783	.8911	.6857	.0277

MACH NO = 20.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0135	.9075	1.2101	-.0882	1.0311	1.0154
.9710	.0141	.8695	1.0358	-.0172	1.0061	.9896
1.1754	.0148	.8184	.9795	.0677	.9761	.9557
1.5025	.0155	.7470	.7353	.1801	.9365	.9058
1.8966	.0160	.6731	.6411	.2900	.8977	.8521
2.3611	.0161	.5999	.5793	.3957	.8605	.7965
2.8972	.0161	.5299	.5301	.4946	.8256	.7408
3.5032	.0158	.4654	.5144	.5839	.7941	.6864
4.1751	.0154	.4075	.5018	.6603	.7671	.6348
4.9062	.0151	.3569	.4986	.7218	.7455	.5868
5.6881	.0149	.3134	.5026	.7680	.7292	.5429
6.5025	.0148	.2851	.5090	.7933	.7202	.5127
7.1518	.0147	.2527	.5208	.8160	.7122	.4762
10.0618	.0151	.1793	.5699	.8280	.7080	.3827
12.8220	.0153	.1399	.6147	.7969	.7190	.3226
15.5493	.0150	.1158	.6539	.7579	.7327	.2793
18.1722	.0203	.1005	.6786	.7223	.7453	.2473
20.6631	.0223	.0906	.7000	.6914	.7562	.2231
23.0214	.0248	.0840	.7168	.6654	.7653	.2041
25.2703	.0275	.0796	.7296	.6453	.7724	.1889
27.6229	.0303	.0763	.7393	.6307	.7776	.1751
29.7592	.0328	.0743	.7447	.6241	.7799	.1643
31.9034	.0350	.0729	.7470	.6244	.7798	.1547
34.0984	.0369	.0720	.7462	.6315	.7773	.1460
36.3917	.0384	.0714	.7425	.6456	.7723	.1378
38.8328	.0394	.0710	.7360	.6670	.7648	.1301
41.6822	.0398	.0708	.7258	.6978	.7539	.1221
44.5188	.0395	.0706	.7143	.7320	.7419	.1151
47.5576	.0390	.0704	.7015	.7694	.7287	.1084
50.7917	.0381	.0703	.6888	.8068	.7155	.1021
54.2088	.0372	.0701	.5776	.8405	.7036	.0962
57.8106	.0364	.0699	.5687	.8680	.6939	.0906
61.6500	.0357	.0697	.5626	.8881	.6868	.0854
66.1625	.0351	.0694	.5592	.9015	.6821	.0800
70.7479	.0348	.0692	.5586	.9071	.6801	.0751
75.8487	.0346	.0689	.5599	.9076	.6799	.0704
81.6033	.0346	.0687	.5623	.9047	.6810	.0657
88.1888	.0346	.0685	.5654	.9003	.6825	.0610
95.8319	.0347	.0683	.5684	.8958	.6841	.0564
104.821	.0347	.0682	.5710	.8925	.6853	.0518
116.5128	.0347	.0681	.5730	.8908	.6859	.0468
129.6238	.0347	.0680	.5743	.8907	.6859	.0422
145.6004	.0346	.0679	.5753	.8916	.6856	.0377
164.1778	.0345	.0678	.5762	.8923	.6853	.0336
200.3417	.0344	.0678	.5780	.8920	.6854	.0277

MACH NO = 25.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0135	.0055	1.2101	-.0882	1.0311	1.0154
.9714	.0141	.0077	1.0364	-.0175	1.0062	.9897
1.1740	.0148	.0178	.8803	.0672	.9763	.9559
1.4996	.0155	.0466	.7359	.1794	.9367	.9062
1.8919	.0159	.0729	.6415	.2892	.8980	.8527
2.3539	.0161	.0999	.5792	.3949	.8607	.7973
2.8870	.0160	.1301	.5387	.4939	.8258	.7418
3.4892	.0157	.1657	.5135	.5836	.7942	.6876
4.1565	.0153	.2079	.5003	.6606	.7670	.6361
4.8818	.0150	.2573	.4966	.7229	.7451	.5883
5.6559	.0148	.3138	.5000	.7700	.7284	.5445
6.2654	.0146	.2856	.5061	.7960	.7193	.5144
7.1056	.0145	.2531	.5174	.8197	.7109	.4780
10.2021	.0149	.1754	.5694	.8327	.7064	.3791
13.1356	.0151	.1351	.6168	.7978	.7187	.3169
16.2110	.0180	.1097	.6565	.7529	.7345	.2705
18.9085	.0202	.0955	.6841	.7156	.7476	.2396
21.6159	.0228	.0858	.7073	.6802	.7601	.2150
23.9624	.0255	.0800	.7241	.6527	.7698	.1975
26.3506	.0285	.0759	.7375	.6305	.7777	.1823
28.4842	.0313	.0734	.7459	.6173	.7823	.1706
30.7421	.0340	.0715	.7510	.6114	.7844	.1598
32.8481	.0363	.0706	.7523	.6135	.7837	.1508
35.1803	.0383	.0698	.7502	.6240	.7800	.1420
37.4585	.0397	.0694	.7450	.6415	.7738	.1343
40.0778	.0405	.0691	.7362	.6687	.7642	.1265
42.6811	.0405	.0690	.7251	.7012	.7527	.1195
45.6706	.0400	.0689	.7111	.7418	.7384	.1125
48.8296	.0392	.0688	.6972	.7820	.7242	.1062
51.9974	.0380	.0686	.6830	.8236	.7095	.0999
55.2826	.0370	.0685	.6718	.8569	.6978	.0945
59.0078	.0360	.0682	.6629	.8845	.6881	.0889
62.6953	.0354	.0680	.6580	.9015	.6821	.0841
66.9865	.0349	.0678	.6559	.9110	.6787	.0790
71.3464	.0347	.0675	.6565	.9132	.6780	.0745
76.5509	.0346	.0673	.6590	.9104	.6789	.0697
81.9779	.0345	.0671	.6622	.9052	.6808	.0654
88.6353	.0347	.0668	.6660	.8989	.6830	.0607
95.7765	.0348	.0667	.6691	.8939	.6848	.0564
104.0980	.0349	.0666	.6716	.8906	.6859	.0521
114.7600	.0349	.0664	.6733	.8894	.6864	.0475
126.7112	.0348	.0664	.6743	.8900	.6861	.0431
142.4901	.0347	.0663	.6751	.8916	.6856	.0385
160.6819	.0346	.0662	.6758	.8928	.6852	.0343
200.6052	.0345	.0661	.6778	.8924	.6853	.0276

MACH NO = 30.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/PB
.8264	.0135	.9059	1.2101	-.0882	1.0311	1.0154
.9710	.0141	.8671	1.0367	-.0176	1.0062	.9898
1.1732	.0147	.8173	.8807	.0669	.9764	.9561
1.4980	.0154	.7463	.7363	.1790	.9369	.9064
1.8802	.0159	.6728	.6417	.2887	.8982	.8531
2.3499	.0160	.5999	.5704	.3943	.8610	.7978
2.8811	.0159	.5302	.5385	.4935	.8260	.7423
3.4811	.0156	.4659	.5131	.5833	.7943	.6883
4.1456	.0153	.4081	.4996	.6607	.7670	.6369
4.8677	.0150	.3576	.4955	.7234	.7449	.5891
5.6390	.0147	.3141	.4986	.7711	.7281	.5454
6.2441	.0145	.2858	.5044	.7975	.7188	.5154
7.0792	.0144	.2534	.5154	.8217	.7102	.4791
10.3758	.0148	.1717	.5707	.8345	.7057	.3747
13.4975	.0161	.1306	.6205	.7955	.7195	.3106
16.7139	.0182	.1058	.6609	.7484	.7361	.2641
19.9125	.0204	.0920	.6803	.7090	.7500	.2337
22.0978	.0221	.0834	.7112	.6739	.7623	.2112
24.6626	.0242	.0776	.7295	.6430	.7733	.1928
26.9139	.0291	.0741	.7419	.6219	.7807	.1791
29.2310	.0322	.0717	.7505	.6094	.7854	.1689
31.3567	.0340	.0702	.7546	.6042	.7869	.1570
33.4974	.0372	.0692	.7552	.6079	.7856	.1482
35.8670	.0392	.0686	.7520	.6208	.7811	.1396
38.2040	.0404	.0683	.7457	.6412	.7739	.1320
40.8743	.0410	.0681	.7354	.6721	.7630	.1243
43.5357	.0409	.0680	.7229	.7084	.7502	.1174
46.3570	.0402	.0680	.7085	.7495	.7357	.1109
49.5601	.0390	.0679	.6926	.7953	.7196	.1044
52.7126	.0379	.0677	.6789	.8352	.7055	.0987
56.0122	.0368	.0676	.6677	.8686	.6937	.0933
59.7315	.0358	.0673	.6593	.8947	.6845	.0879
63.4376	.0352	.0671	.6551	.9097	.6792	.0832
67.7298	.0348	.0668	.6541	.9164	.6768	.0782
72.1106	.0346	.0666	.6557	.9160	.6770	.0738
76.9407	.0346	.0663	.6588	.9113	.6786	.0694
82.7368	.0347	.0661	.6628	.9041	.6812	.0648
88.8743	.0348	.0659	.6666	.8975	.6835	.0606
96.4443	.0349	.0658	.6699	.8920	.6854	.0560
104.6989	.0350	.0656	.6722	.8891	.6864	.0518
114.4436	.0350	.0655	.6736	.8886	.6866	.0476
127.0603	.0349	.0655	.6744	.8899	.6862	.0430
141.5337	.0348	.0654	.6750	.8917	.6855	.0388
160.9204	.0347	.0653	.6757	.8932	.6850	.0342
201.6585	.0345	.0652	.6778	.8925	.6852	.0275

MACH NO = 3.50 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0134	1.0310	1.3492	-.1340	1.0718	1.0353
.8768	.0143	.9763	1.1518	-.0654	1.0356	.9977
1.0381	.0154	.9190	1.0076	-.0018	1.0010	.9565
1.2268	.0164	.8604	.8951	.0587	.9685	.9124
1.5248	.0178	.7823	.7958	.1324	.9291	.8504
1.7845	.0187	.7257	.7456	.1822	.9023	.8029
2.1719	.0198	.6563	.7019	.2400	.8714	.7411
2.4942	.0205	.6091	.6804	.2776	.8512	.6966
2.9667	.0213	.5530	.6640	.3191	.8290	.6401
3.3524	.0219	.5153	.6576	.3446	.8153	.6004
3.9082	.0227	.4739	.6549	.3716	.8008	.5511
4.3558	.0233	.4468	.6558	.3875	.7923	.5169
4.0932	.0241	.4161	.6591	.4043	.7833	.4750
6.0364	.0253	.3800	.6662	.4224	.7736	.4193
7.1861	.0265	.3535	.6736	.4355	.7656	.3714
8.4476	.0275	.3341	.6801	.4459	.7611	.3299
9.8299	.0284	.3199	.6852	.4550	.7562	.2940
11.3469	.0292	.3096	.6890	.4634	.7516	.2626
13.0153	.0299	.3020	.6917	.4716	.7473	.2350
14.8199	.0304	.2966	.6935	.4794	.7431	.2110
16.7667	.0308	.2927	.6945	.4869	.7391	.1901
18.8736	.0311	.2898	.6951	.4940	.7353	.1717
21.1612	.0313	.2877	.6954	.5006	.7317	.1553
23.6527	.0314	.2861	.6957	.5066	.7285	.1407
26.3727	.0315	.2849	.6959	.5119	.7257	.1277
29.3474	.0315	.2840	.6962	.5165	.7232	.1159
32.6048	.0315	.2834	.6966	.5204	.7211	.1052
36.1752	.0315	.2828	.6972	.5235	.7194	.0956
40.7806	.0315	.2824	.6981	.5264	.7179	.0855
45.1464	.0315	.2821	.6989	.5284	.7168	.0777
49.9403	.0315	.2818	.6998	.5299	.7160	.0707
55.2056	.0315	.2816	.7007	.5310	.7154	.0643
60.9900	.0314	.2815	.7016	.5318	.7150	.0584
67.3455	.0314	.2814	.7026	.5324	.7147	.0532
74.3293	.0314	.2813	.7035	.5328	.7145	.0483
82.0042	.0314	.2812	.7044	.5330	.7143	.0440
90.4332	.0314	.2812	.7052	.5332	.7143	.0400
99.7098	.0314	.2811	.7060	.5332	.7142	.0364
109.8994	.0313	.2811	.7068	.5332	.7142	.0331
121.0992	.0313	.2811	.7075	.5332	.7143	.0301
133.4098	.0313	.2811	.7082	.5331	.7143	.0274
146.9415	.0313	.2810	.7088	.5331	.7143	.0249
161.8156	.0313	.2810	.7093	.5330	.7144	.0227
178.1656	.0313	.2810	.7098	.5329	.7144	.0206
202.5165	.0313	.2810	.7104	.5328	.7145	.0182

MACH NO = 5.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0133	.9837	1.3492	-.1340	1.0718	1.0353
.8732	.0140	.9307	1.1551	-.0676	1.0362	.9987
1.0375	.0149	.8722	1.0008	-.0004	1.0002	.9566
1.2898	.0159	.7947	.8613	.0804	.9569	.8985
1.5806	.0168	.7201	.7718	.1515	.9188	.8397
1.9104	.0175	.6503	.7142	.2134	.8857	.7817
2.1829	.0179	.6019	.6856	.2538	.8640	.7395
2.5806	.0184	.5429	.6604	.3004	.8390	.6855
3.0159	.0189	.4908	.6476	.3373	.8193	.6347
3.4855	.0195	.4457	.6431	.3656	.8041	.5878
3.9864	.0200	.4071	.6446	.3859	.7932	.5448
4.3805	.0205	.3822	.6482	.3969	.7873	.5152
4.9281	.0212	.3537	.6550	.4072	.7818	.4790
6.0897	.0228	.3101	.6711	.4174	.7763	.4168
7.1697	.0243	.2833	.6844	.4208	.7745	.3720
8.3060	.0259	.2640	.6953	.4228	.7734	.3341
9.5019	.0273	.2501	.7034	.4253	.7721	.3018
10.9511	.0288	.2389	.7092	.4301	.7695	.2701
12.3041	.0300	.2319	.7117	.4363	.7662	.2461
13.7534	.0309	.2267	.7123	.4443	.7619	.2246
15.3223	.0316	.2230	.7116	.4535	.7570	.2052
17.3016	.0322	.2198	.7096	.4650	.7508	.1851
19.2387	.0325	.2178	.7071	.4757	.7451	.1689
21.4138	.0327	.2163	.7041	.4870	.7390	.1537
24.2561	.0327	.2150	.7005	.4998	.7322	.1376
27.0333	.0326	.2141	.6980	.5096	.7269	.1248
30.0712	.0325	.2134	.6965	.5172	.7228	.1133
33.4000	.0324	.2129	.6958	.5229	.7198	.1029
37.6018	.0323	.2124	.6958	.5275	.7173	.0922
41.6648	.0322	.2121	.6963	.5303	.7158	.0838
46.1269	.0322	.2118	.6971	.5321	.7148	.0762
51.0286	.0321	.2116	.6982	.5333	.7142	.0693
57.2257	.0321	.2114	.6995	.5341	.7138	.0621
63.2237	.0320	.2112	.7007	.5344	.7136	.0565
69.8153	.0320	.2111	.7018	.5345	.7136	.0513
78.1519	.0320	.2110	.7031	.5344	.7136	.0461
86.2229	.0320	.2110	.7042	.5343	.7137	.0419
95.0945	.0319	.2109	.7051	.5342	.7137	.0381
104.8467	.0319	.2109	.7060	.5340	.7138	.0346
117.1833	.0319	.2109	.7070	.5338	.7140	.0311
129.1294	.0319	.2108	.7077	.5336	.7140	.0283
142.2624	.0319	.2108	.7084	.5334	.7141	.0257
156.7007	.0319	.2108	.7090	.5333	.7142	.0234
174.9672	.0319	.2108	.7096	.5331	.7143	.0210
200.7665	.0319	.2108	.7103	.5330	.7144	.0183

MACH NO = 10.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0131	.9468	1.3492	-.1340	1.0718	1.0353
.9069	.0139	.8816	1.1162	-.0522	1.0279	.9898
1.0597	.0145	.8283	.9817	.0092	.9951	.9512
1.2924	.0152	.7571	.8536	.0849	.9545	.8980
1.6301	.0158	.6710	.7495	.1695	.9091	.8305
1.9369	.0161	.6066	.6961	.2288	.8774	.7774
2.2747	.0163	.5473	.6604	.2805	.8497	.7263
2.6417	.0165	.4937	.6379	.3242	.8263	.6779
3.1357	.0166	.4354	.6240	.3668	.8035	.6221
3.5537	.0168	.3955	.6213	.3914	.7903	.5816
3.9868	.0171	.3612	.6238	.4085	.7811	.5448
4.4308	.0175	.3319	.6300	.4193	.7753	.5116
4.9955	.0181	.3015	.6409	.4259	.7718	.4749
6.2518	.0198	.2529	.6695	.4230	.7733	.4094
7.5025	.0219	.2218	.6964	.4100	.7803	.3600
8.7325	.0243	.2016	.7171	.3976	.7869	.3218
9.9370	.0267	.1884	.7310	.3896	.7912	.2915
11.1208	.0290	.1797	.7400	.3858	.7932	.2669
12.2972	.0311	.1739	.7454	.3853	.7935	.2462
13.4859	.0329	.1701	.7477	.3883	.7919	.2283
14.7099	.0344	.1676	.7467	.3956	.7880	.2124
15.8748	.0354	.1661	.7429	.4064	.7822	.1992
17.2394	.0359	.1650	.7358	.4227	.7735	.1856
18.7228	.0360	.1642	.7262	.4431	.7626	.1729
20.3597	.0357	.1636	.7154	.4656	.7505	.1607
22.1754	.0351	.1631	.7049	.4877	.7386	.1490
24.1838	.0345	.1625	.6962	.5069	.7283	.1380
26.4091	.0339	.1619	.6901	.5217	.7204	.1275
28.9098	.0335	.1613	.6869	.5316	.7151	.1175
31.7673	.0332	.1607	.6863	.5370	.7122	.1078
35.0840	.0331	.1602	.6877	.5388	.7113	.0983
38.9888	.0330	.1597	.6903	.5384	.7115	.0892
43.6438	.0330	.1593	.6932	.5372	.7121	.0802
49.2517	.0330	.1590	.6960	.5360	.7127	.0716
55.8143	.0329	.1587	.6984	.5354	.7131	.0636
63.1844	.0329	.1586	.7002	.5351	.7132	.0565
71.4657	.0329	.1584	.7018	.5350	.7133	.0502
80.7745	.0328	.1584	.7032	.5349	.7134	.0446
91.2408	.0328	.1583	.7045	.5347	.7135	.0397
103.0104	.0328	.1582	.7057	.5344	.7136	.0352
116.2470	.0328	.1582	.7068	.5340	.7138	.0313
131.1345	.0328	.1582	.7077	.5338	.7140	.0279
147.8797	.0328	.1582	.7086	.5335	.7141	.0248
166.7153	.0328	.1582	.7093	.5333	.7142	.0220
200.5545	.0328	.1581	.7103	.5331	.7143	.0183

MACH NO = 15.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.74..	.0131	.9397	1.3492	-.1340	1.0718	1.0353
.9031	.0139	.8760	1.1201	-.0537	1.0288	.9908
1.0521	.0144	.8238	.9865	.0067	.9964	.9531
1.3407	.0151	.7366	.8327	.0995	.9467	.8876
1.6077	.0155	.6691	.7522	.1662	.9109	.8347
1.9058	.0158	.6053	.6971	.2259	.8790	.7825
2.3192	.0160	.5327	.6526	.2900	.8446	.7201
2.6801	.0160	.4809	.6312	.3327	.8217	.6732
3.0633	.0161	.4351	.6194	.3671	.8033	.6297
3.5665	.0163	.3861	.6149	.3986	.7864	.5804
3.9829	.0165	.3529	.6172	.4155	.7773	.5451
4.4067	.0167	.3245	.6232	.4262	.7716	.5133
4.9419	.0172	.2949	.6339	.4325	.7682	.4781
6.3296	.0190	.2405	.6683	.4261	.7717	.4060
7.6763	.0214	.2077	.7003	.4073	.7817	.3541
9.0565	.0243	.1862	.7247	.3903	.7908	.3131
10.2707	.0269	.1739	.7399	.3796	.7966	.2841
11.4319	.0296	.1661	.7509	.3717	.8008	.2611
12.5613	.0321	.1611	.7580	.3677	.8030	.2419
13.7700	.0345	.1579	.7609	.3693	.8021	.2244
14.9114	.0361	.1561	.7590	.3773	.7978	.2100
16.0967	.0372	.1551	.7529	.3915	.7902	.1968
17.3530	.0375	.1545	.7431	.4114	.7795	.1846
18.8162	.0373	.1541	.7297	.4377	.7654	.1721
20.2939	.0366	.1538	.7160	.4644	.7511	.1612
21.8933	.0358	.1534	.7028	.4904	.7372	.1507
23.6198	.0349	.1530	.6919	.5127	.7252	.1409
25.4935	.0341	.1524	.6845	.5293	.7164	.1316
27.7238	.0336	.1517	.6809	.5397	.7108	.1220
30.0476	.0333	.1511	.6811	.5434	.7088	.1134
32.6811	.0332	.1505	.6835	.5430	.7090	.1050
35.7087	.0332	.1500	.6871	.5406	.7103	.0968
39.5380	.0332	.1495	.6911	.5374	.7120	.0880
43.7800	.0333	.1491	.6944	.5353	.7132	.0800
48.8853	.0333	.1489	.6968	.5344	.7136	.0721
55.1159	.0333	.1487	.6985	.5348	.7134	.0644
63.4572	.0332	.1485	.7001	.5354	.7131	.0563
72.5648	.0331	.1483	.7016	.5356	.7130	.0495
82.9099	.0331	.1483	.7032	.5353	.7132	.0435
94.6624	.0331	.1482	.7048	.5347	.7135	.0383
109.1151	.0331	.1481	.7063	.5341	.7138	.0333
124.4394	.0331	.1481	.7074	.5337	.7140	.0293
141.8558	.0331	.1481	.7084	.5333	.7142	.0258
161.6519	.0331	.1481	.7092	.5331	.7143	.0227
201.5107	.0331	.1480	.7103	.5329	.7144	.0183

MACH NO = 20.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0131	.9370	1.3492	-.1340	1.0718	1.0353
.9017	.0138	.8779	1.1216	-.0543	1.0291	.9911
1.0494	.0144	.8221	.9883	.0058	.9969	.9537
1.3357	.0151	.7355	.9343	.0983	.9473	.8888
1.5995	.0155	.6683	.7533	.1650	.9116	.8362
1.9728	.0157	.5897	.6866	.2385	.8722	.7716
2.3033	.0158	.5324	.6523	.2892	.8456	.7223
2.7522	.0159	.4688	.6263	.3417	.8169	.6645
3.1347	.0159	.4246	.6158	.3746	.7992	.6222
3.6345	.0161	.3773	.6125	.4044	.7873	.5743
4.0458	.0162	.3453	.6153	.4203	.7748	.5401
4.4629	.0165	.3180	.6216	.4301	.7695	.5094
4.9875	.0170	.2893	.6325	.4356	.7666	.4753
5.4412	.0189	.2376	.6700	.4263	.7716	.4011
7.9286	.0216	.1991	.7056	.4036	.7837	.3458
9.2289	.0243	.1811	.7281	.3872	.7925	.3086
10.5300	.0273	.1677	.7450	.3740	.7996	.2786
11.6773	.0302	.1606	.7567	.3645	.8047	.2567
12.7859	.0329	.1562	.7640	.3597	.8072	.2385
13.9610	.0353	.1534	.7664	.3617	.8062	.2218
15.0727	.0370	.1520	.7636	.3707	.8013	.2081
16.3093	.0380	.1512	.7557	.3878	.7922	.1947
17.5348	.0382	.1508	.7446	.4097	.7804	.1830
18.9496	.0379	.1506	.7297	.4382	.7652	.1711
20.3700	.0369	.1504	.7146	.4670	.7498	.1606
21.8934	.0359	.1501	.7002	.4948	.7348	.1507
23.6448	.0348	.1496	.6878	.5197	.7215	.1408
25.4146	.0340	.1490	.6805	.5358	.7129	.1320
27.4954	.0335	.1483	.6777	.5447	.7081	.1229
29.6555	.0332	.1477	.6788	.5467	.7070	.1148
32.2622	.0332	.1470	.6823	.5445	.7082	.1062
35.0386	.0332	.1465	.6865	.5408	.7102	.0985
38.2373	.0334	.1460	.6906	.5371	.7122	.0908
42.2714	.0335	.1457	.6942	.5343	.7136	.0827
46.7817	.0335	.1454	.6965	.5336	.7140	.0752
52.6845	.0334	.1452	.6981	.5344	.7136	.0672
59.5270	.0333	.1451	.6992	.5355	.7130	.0598
68.7556	.0332	.1448	.7008	.5361	.7127	.0521
79.0011	.0332	.1447	.7026	.5356	.7130	.0456
90.6972	.0332	.1446	.7044	.5347	.7134	.0399
105.0748	.0332	.1446	.7061	.5339	.7139	.0346
120.4736	.0332	.1445	.7074	.5334	.7142	.0303
139.4041	.0332	.1445	.7084	.5331	.7143	.0262
159.6855	.0332	.1445	.7093	.5329	.7144	.0230
201.4021	.0332	.1445	.7104	.5328	.7145	.0183

MACH NO = 25.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0131	.9360	1.3492	-.1340	1.0718	1.0353
.8985	.0138	.8741	1.1251	-.0557	1.0299	.9920
1.0909	.0145	.8075	.9591	.0210	.9887	.9437
1.3124	.0150	.7408	.8431	.0920	.9507	.8937
1.6304	.0154	.6601	.7456	.1722	.9077	.8304
1.9166	.0157	.5994	.6937	.2292	.8772	.7807
2.3113	.0158	.5300	.6508	.2910	.8440	.7212
2.6553	.0158	.4801	.6293	.3328	.8217	.6762
3.1145	.0158	.4254	.6148	.3744	.7993	.6243
3.4998	.0159	.3877	.6110	.3992	.7861	.5865
3.9967	.0161	.3475	.6133	.4204	.7747	.5440
4.4014	.0163	.3203	.6190	.4307	.7692	.5137
4.9111	.0167	.2917	.6294	.4369	.7659	.4800
6.4254	.0186	.2324	.6688	.4274	.7709	.4018
7.8687	.0213	.1984	.7043	.4045	.7832	.3477
9.3005	.0243	.1774	.7296	.3858	.7933	.3068
10.5443	.0273	.1656	.7464	.3721	.8006	.2783
11.7140	.0303	.1584	.7591	.3612	.8064	.2560
12.8391	.0332	.1541	.7669	.3557	.8094	.2377
13.9491	.0356	.1515	.7692	.3573	.8085	.2220
15.0715	.0374	.1501	.7662	.3666	.8036	.2081
16.3141	.0384	.1494	.7579	.3843	.7940	.1946
17.5521	.0386	.1491	.7459	.4076	.7816	.1828
18.8796	.0381	.1491	.7310	.4358	.7665	.1716
20.3012	.0371	.1488	.7148	.4665	.7500	.1611
21.8191	.0359	.1485	.6992	.4962	.7341	.1512
23.5509	.0348	.1480	.6860	.5224	.7201	.1413
25.3072	.0339	.1474	.6785	.5389	.7112	.1325
27.2287	.0334	.1468	.6761	.5470	.7069	.1240
29.3550	.0332	.1461	.6775	.5493	.7062	.1158
31.7339	.0332	.1455	.6813	.5454	.7077	.1079
34.4268	.0333	.1449	.6860	.5410	.7101	.1001
37.7383	.0334	.1444	.6907	.5365	.7125	.0919
41.3755	.0335	.1441	.6942	.5337	.7140	.0844
45.6965	.0336	.1438	.6965	.5330	.7144	.0769
50.9355	.0335	.1436	.6978	.5340	.7139	.0694
57.4143	.0334	.1434	.6987	.5355	.7130	.0619
66.1784	.0333	.1432	.7002	.5363	.7126	.0541
76.5161	.0332	.1431	.7021	.5358	.7129	.0470
88.4977	.0332	.1430	.7042	.5347	.7135	.0408
102.2718	.0333	.1429	.7060	.5337	.7140	.0355
118.1103	.0333	.1428	.7074	.5331	.7143	.0309
136.3271	.0333	.1428	.7084	.5328	.7144	.0268
158.7872	.0333	.1428	.7094	.5327	.7145	.0231
201.3478	.0333	.1428	.7105	.5326	.7146	.0183

MACH NO = 30.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0131	.9353	1.3492	-.1340	1.0718	1.0353
.8982	.0138	.8735	1.1254	-.0559	1.0299	.9921
1.0901	.0145	.8071	.9596	.0208	.9889	.9439
1.3111	.0150	.7405	.8435	.0917	.9509	.8940
1.6283	.0154	.6599	.7459	.1719	.9079	.8308
1.9893	.0157	.5849	.6835	.2421	.8703	.7690
2.3074	.0157	.5299	.6507	.2908	.8442	.7217
2.7395	.0158	.4685	.6251	.3420	.8167	.6661
3.2026	.0158	.4155	.6128	.3814	.7956	.6152
3.5896	.0159	.3790	.6103	.4045	.7832	.5783
4.0866	.0161	.3403	.6136	.4239	.7728	.5370
4.4903	.0163	.3141	.6199	.4331	.7679	.5075
4.9974	.0167	.2865	.6307	.4381	.7652	.4747
6.5963	.0188	.2266	.6731	.4253	.7721	.3945
8.0143	.0215	.1948	.7073	.4024	.7844	.3431
9.4152	.0245	.1750	.7315	.3842	.7941	.3039
10.6290	.0274	.1639	.7481	.3702	.8016	.2766
11.8434	.0307	.1567	.7614	.3585	.8079	.2538
12.9370	.0335	.1527	.7687	.3533	.8106	.2362
14.0878	.0360	.1502	.7706	.3557	.8094	.2202
15.1797	.0377	.1490	.7671	.3656	.8041	.2068
16.3890	.0387	.1484	.7584	.3837	.7944	.1938
17.6758	.0388	.1482	.7454	.4089	.7809	.1817
18.9674	.0382	.1481	.7303	.4373	.7657	.1710
20.4398	.0371	.1480	.7128	.4701	.7481	.1601
21.9105	.0359	.1477	.6973	.4995	.7323	.1506
23.5832	.0347	.1472	.6844	.5251	.7186	.1411
25.2754	.0339	.1466	.6773	.5410	.7101	.1326
27.2495	.0333	.1459	.6750	.5487	.7060	.1239
29.2955	.0332	.1452	.6769	.5492	.7057	.1160
31.7338	.0332	.1445	.6812	.5456	.7076	.1079
34.3154	.0333	.1440	.6860	.5408	.7102	.1004
37.4660	.0335	.1435	.6907	.5361	.7127	.0925
41.1454	.0336	.1432	.6943	.5333	.7142	.0848
45.2313	.0336	.1429	.6965	.5327	.7145	.0776
50.5109	.0335	.1427	.6977	.5339	.7139	.0699
56.6337	.0334	.1425	.6985	.5355	.7130	.0627
64.8724	.0333	.1423	.6998	.5365	.7125	.0551
74.7025	.0333	.1422	.7017	.5360	.7128	.0481
87.1166	.0333	.1421	.7041	.5347	.7135	.0415
100.5361	.0333	.1420	.7059	.5336	.7141	.0361
117.0489	.0333	.1419	.7074	.5329	.7144	.0311
134.9088	.0333	.1419	.7085	.5326	.7146	.0271
156.8958	.0333	.1419	.7094	.5325	.7146	.0234
200.3492	.0333	.1419	.7106	.5324	.7147	.0184

MACH NO = 3.50 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	1.0720	1.5198	-.1820	1.1325	1.0642
.8043	.0139	.9959	1.2623	-.1063	1.0773	1.0071
.9297	.0149	.9405	1.1264	-.0566	1.0412	.9628
1.1611	.0163	.8552	.9768	.0120	.9913	.8906
1.3664	.0174	.7939	.9006	.0567	.9587	.8350
1.5835	.0183	.7386	.8482	.0943	.9313	.7820
1.8924	.0194	.6781	.8059	.1328	.9033	.7199
2.1529	.0201	.6352	.7822	.1580	.8850	.6740
2.4316	.0208	.5998	.7663	.1792	.8699	.6308
2.8028	.0217	.5595	.7551	.1995	.8548	.5813
3.1163	.0224	.5330	.7498	.2126	.8453	.5451
3.4435	.0231	.5103	.7474	.2226	.8380	.5119
3.8701	.0240	.4869	.7466	.2325	.8308	.4742
4.4987	.0252	.4615	.7477	.2428	.8233	.4278
5.2628	.0264	.4406	.7497	.2518	.8167	.3823
6.0825	.0276	.4258	.7514	.2594	.8112	.3432
6.8542	.0285	.4165	.7521	.2660	.8064	.3130
7.8115	.0297	.4089	.7513	.2742	.8004	.2822
8.8671	.0298	.4035	.7490	.2833	.7938	.2546
9.8899	.0301	.4001	.7464	.2917	.7877	.2326
11.1982	.0304	.3972	.7433	.3009	.7810	.2094
12.6886	.0305	.3951	.7406	.3094	.7748	.1880
14.1747	.0305	.3938	.7387	.3161	.7699	.1707
16.1248	.0304	.3926	.7370	.3228	.7651	.1522
18.0937	.0303	.3919	.7363	.3274	.7616	.1373
20.6542	.0303	.3912	.7362	.3315	.7587	.1217
23.5318	.0302	.3907	.7368	.3342	.7567	.1079
26.3424	.0301	.3904	.7378	.3357	.7556	.0972
29.9302	.0301	.3901	.7391	.3368	.7548	.0863
33.9681	.0300	.3899	.7405	.3374	.7544	.0765
37.9153	.0300	.3898	.7418	.3376	.7543	.0690
42.9572	.0300	.3897	.7432	.3376	.7542	.0612
48.6342	.0299	.3896	.7446	.3376	.7543	.0543
54.1856	.0299	.3896	.7456	.3374	.7544	.0490
61.2783	.0299	.3896	.7468	.3373	.7545	.0435
68.2150	.0299	.3895	.7477	.3371	.7546	.0392
77.0783	.0299	.3895	.7486	.3370	.7547	.0348
87.0608	.0299	.3895	.7494	.3369	.7548	.0309
96.8242	.0299	.3895	.7500	.3368	.7549	.0278
109.3004	.0299	.3895	.7506	.3367	.7549	.0247
123.3525	.0299	.3895	.7512	.3366	.7550	.0219
137.0967	.0299	.3895	.7516	.3366	.7550	.0198
154.6631	.0299	.3895	.7520	.3365	.7550	.0175
174.4423	.0299	.3895	.7523	.3365	.7550	.0156
202.7235	.0299	.3895	.7527	.3365	.7550	.0134

MACH NO = 5.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6590	.0127	1.0245	1.5198	-.1820	1.1325	1.0642
.7886	.0135	.9559	1.2828	-.1129	1.0922	1.0129
.9389	.0145	.8882	1.1148	-.0517	1.0377	.9597
1.1498	.0156	.8086	.9753	.0127	.9907	.8939
1.3855	.0165	.7358	.8853	.0659	.9520	.8302
1.6446	.0172	.6708	.8271	.1095	.9203	.7699
1.9245	.0180	.6138	.7900	.1443	.8950	.7139
2.2224	.0186	.5647	.7670	.1716	.8751	.6626
2.5394	.0193	.5224	.7542	.1921	.8602	.6157
2.8681	.0201	.4869	.7484	.2068	.8494	.5734
3.2085	.0209	.4573	.7474	.2170	.8421	.5353
3.4867	.0216	.4373	.7487	.2224	.8381	.5078
3.8401	.0225	.4164	.7518	.2271	.8347	.4767
4.5634	.0243	.3851	.7600	.2319	.8312	.4235
5.2334	.0260	.3657	.7667	.2344	.8294	.3839
6.0033	.0278	.3510	.7711	.2381	.8267	.3466
6.7253	.0291	.3419	.7717	.2439	.8225	.3177
7.4840	.0301	.3356	.7696	.2518	.8167	.2920
8.3848	.0309	.3307	.7656	.2619	.8093	.2665
9.2638	.0314	.3276	.7610	.2718	.8021	.2456
10.3400	.0317	.3252	.7550	.2838	.7934	.2240
11.4231	.0317	.3236	.7490	.2951	.7852	.2059
12.6431	.0316	.3224	.7472	.3062	.7771	.1886
14.2000	.0314	.3212	.7380	.3170	.7692	.1704
15.8274	.0312	.3204	.7349	.3247	.7636	.1548
17.9493	.0310	.3195	.7332	.3310	.7591	.1383
20.2054	.0308	.3188	.7331	.3347	.7563	.1242
22.8600	.0307	.3182	.7341	.3370	.7547	.1109
26.2779	.0306	.3177	.7359	.3380	.7539	.0974
29.7469	.0306	.3174	.7378	.3383	.7538	.0868
33.6355	.0306	.3171	.7397	.3382	.7538	.0773
38.5112	.0305	.3169	.7417	.3380	.7540	.0679
43.4629	.0305	.3168	.7433	.3377	.7543	.0605
49.6742	.0305	.3168	.7448	.3375	.7543	.0532
55.9845	.0305	.3167	.7460	.3373	.7544	.0474
63.0543	.0305	.3167	.7471	.3372	.7546	.0423
71.9483	.0305	.3167	.7481	.3370	.7547	.0372
80.9765	.0305	.3166	.7489	.3369	.7547	.0331
92.3066	.0305	.3166	.7497	.3368	.7548	.0292
103.8214	.0305	.3166	.7503	.3367	.7549	.0260
116.7429	.0305	.3166	.7509	.3367	.7549	.0232
132.9598	.0305	.3166	.7514	.3366	.7550	.0204
149.4415	.0305	.3166	.7518	.3366	.7550	.0182
170.1269	.0305	.3166	.7522	.3366	.7550	.0160
201.3004	.0305	.3166	.7527	.3366	.7550	.0135

MACH NO = 10.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	.9874	1.5198	-.1820	1.1325	1.0642
.8080	.0136	.9094	1.2552	-.1037	1.0755	1.0058
.9834	.0144	.8322	1.0746	-.0347	1.0252	.9451
1.1812	.0151	.7589	.9536	.0243	.9823	.8848
1.4461	.0157	.6783	.8589	.0832	.9394	.8153
1.6871	.0162	.6184	.8082	.1231	.9104	.7609
1.9953	.0167	.5559	.7702	.1608	.8830	.7010
2.2643	.0171	.5114	.7516	.1845	.8657	.6560
2.5983	.0177	.4660	.7411	.2044	.8512	.6076
2.8804	.0183	.4346	.7389	.2150	.8435	.5719
3.2206	.0191	.4033	.7418	.2220	.8384	.5341
3.5030	.0198	.3818	.7470	.2244	.8367	.5063
3.8391	.0208	.3606	.7550	.2242	.8368	.4768
4.5523	.0234	.3274	.7736	.2186	.8409	.4242
5.2949	.0261	.3052	.7875	.2141	.8442	.3806
5.9650	.0283	.2922	.7935	.2145	.8438	.3483
6.6751	.0304	.2834	.7963	.2172	.8419	.3195
7.3358	.0321	.2783	.7952	.2217	.8386	.2967
8.0652	.0334	.2751	.7921	.2305	.8322	.2750
8.7743	.0340	.2733	.7843	.2430	.8231	.2568
9.5294	.0342	.2722	.7735	.2589	.8115	.2399
10.4132	.0338	.2714	.7599	.2784	.7973	.2227
11.3181	.0333	.2706	.7475	.2964	.7842	.2075
12.3976	.0327	.2698	.7365	.3133	.7719	.1918
13.5074	.0321	.2689	.7294	.3254	.7631	.1780
14.8429	.0317	.2679	.7253	.3340	.7568	.1639
16.2488	.0314	.2669	.7247	.3381	.7539	.1512
17.8590	.0313	.2660	.7252	.3395	.7529	.1389
19.8886	.0313	.2652	.7291	.3393	.7530	.1260
22.1325	.0313	.2645	.7320	.3387	.7535	.1142
25.0628	.0313	.2639	.7350	.3381	.7539	.1018
28.4131	.0312	.2635	.7374	.3379	.7540	.0906
32.9142	.0312	.2631	.7397	.3378	.7541	.0789
38.0127	.0312	.2629	.7418	.3376	.7542	.0688
43.8568	.0312	.2628	.7437	.3373	.7545	.0600
51.1009	.0312	.2627	.7454	.3370	.7546	.0518
58.8437	.0312	.2626	.7467	.3369	.7548	.0452
68.4468	.0312	.2626	.7479	.3368	.7548	.0390
78.7155	.0312	.2626	.7489	.3367	.7549	.0341
90.4815	.0312	.2626	.7497	.3367	.7549	.0297
105.0791	.0312	.2625	.7505	.3366	.7550	.0257
120.6912	.0312	.2625	.7511	.3366	.7550	.0224
140.0613	.0312	.2625	.7516	.3366	.7550	.0194
160.7780	.0312	.2625	.7521	.3365	.7550	.0169
200.8182	.0312	.2625	.7527	.3365	.7550	.0136

MACH NO = 15.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	.9803	1.5198	-.1820	1.1325	1.0642
.8046	.0135	.9039	1.2595	-.1051	1.0765	1.0070
.9750	.0143	.8279	1.0799	-.0369	1.0269	.9475
1.2100	.0150	.7417	.9388	.0325	.9764	.8767
1.4272	.0155	.6757	.8618	.0808	.9412	.8199
1.7103	.0160	.6050	.8016	.1282	.9067	.7561
2.0123	.0164	.5442	.7654	.1648	.8801	.6980
2.3280	.0169	.4928	.7452	.1917	.8605	.6462
2.5975	.0173	.4567	.7377	.2070	.8493	.6077
2.9233	.0179	.4204	.7362	.2186	.8409	.5668
3.2480	.0187	.3908	.7403	.2241	.8369	.5312
3.5159	.0194	.3704	.7463	.2253	.8360	.5051
3.8325	.0204	.3501	.7553	.2238	.8371	.4773
4.5963	.0233	.3145	.7780	.2150	.8435	.4214
5.3656	.0263	.2919	.7926	.2097	.8474	.3769
6.0516	.0287	.2791	.7996	.2089	.8479	.3445
6.7152	.0310	.2714	.8040	.2092	.8477	.3180
7.3723	.0330	.2669	.8046	.2129	.8450	.2956
8.0394	.0343	.2643	.7999	.2218	.8385	.2758
8.7323	.0351	.2630	.7902	.2362	.8281	.2578
9.5190	.0349	.2622	.7759	.2561	.8135	.2401
10.3162	.0344	.2617	.7606	.2772	.7982	.2245
11.1861	.0336	.2611	.7459	.2978	.7832	.2096
12.1330	.0328	.2603	.7336	.3159	.7700	.1955
13.1661	.0321	.2594	.7251	.3295	.7601	.1821
14.3857	.0317	.2583	.7212	.3378	.7541	.1685
15.6670	.0315	.2572	.7217	.3405	.7521	.1562
17.1148	.0314	.2562	.7244	.3404	.7522	.1443
18.7734	.0314	.2554	.7277	.3393	.7530	.1327
20.7056	.0315	.2547	.7310	.3382	.7538	.1214
23.0030	.0315	.2541	.7339	.3374	.7544	.1102
26.0048	.0315	.2536	.7362	.3375	.7543	.0984
29.5285	.0314	.2531	.7379	.3380	.7540	.0874
33.9860	.0314	.2529	.7400	.3380	.7539	.0765
39.6330	.0314	.2526	.7425	.3374	.7544	.0661
46.2854	.0314	.2524	.7448	.3366	.7550	.0570
54.5454	.0314	.2523	.7466	.3362	.7552	.0487
63.5643	.0314	.2523	.7478	.3362	.7553	.0420
74.0182	.0314	.2522	.7488	.3363	.7552	.0362
86.1393	.0315	.2522	.7497	.3364	.7551	.0312
100.1947	.0315	.2522	.7504	.3364	.7551	.0269
116.4936	.0315	.2522	.7511	.3364	.7551	.0232
136.7569	.0315	.2522	.7517	.3364	.7551	.0198
158.8929	.0315	.2522	.7521	.3365	.7551	.0171
201.8928	.0315	.2522	.7527	.3365	.7551	.0135

MACH NO = 20.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	.9776	1.5198	-.1820	1.1325	1.0642
.8033	.0135	.9018	1.2611	-.1057	1.0769	1.0075
.9732	.0142	.8263	1.0818	-.0378	1.0275	.9484
1.2052	.0150	.7405	.9405	.0315	.9771	.8781
1.4202	.0154	.6748	.8630	.0799	.9418	.8216
1.7005	.0159	.6042	.8022	.1275	.9072	.7581
1.9997	.0163	.5435	.7654	.1642	.8804	.7003
2.3106	.0167	.4921	.7447	.1914	.8607	.6488
2.5761	.0171	.4559	.7368	.2070	.8494	.6106
2.8968	.0177	.4196	.7349	.2188	.8407	.5700
3.2158	.0185	.3899	.7389	.2244	.8366	.5346
3.4785	.0192	.3693	.7449	.2257	.8357	.5086
3.7886	.0202	.3489	.7540	.2241	.8368	.4810
4.5816	.0232	.3110	.7788	.2140	.8442	.4223
5.3239	.0261	.2897	.7935	.2084	.8483	.3791
6.0238	.0287	.2754	.8015	.2066	.8496	.3457
6.6948	.0312	.2675	.8070	.2059	.8501	.3188
7.3549	.0333	.2630	.8080	.2091	.8478	.2961
8.0222	.0347	.2606	.8029	.2183	.8411	.2762
8.7135	.0354	.2594	.7925	.2335	.8300	.2583
9.4462	.0353	.2588	.7780	.2534	.8155	.2416
10.2369	.0347	.2584	.7616	.2757	.7993	.2259
11.0952	.0338	.2578	.7456	.2978	.7832	.2110
12.0251	.0329	.2571	.7321	.3173	.7690	.1970
13.0365	.0321	.2561	.7230	.3316	.7586	.1837
14.1493	.0316	.2550	.7194	.3393	.7530	.1709
15.3871	.0314	.2538	.7203	.3415	.7514	.1587
16.7779	.0314	.2528	.7235	.3407	.7520	.1469
18.3534	.0315	.2519	.7273	.3391	.7531	.1355
20.1864	.0315	.2512	.7307	.3377	.7542	.1243
22.3410	.0316	.2506	.7336	.3370	.7547	.1132
24.9477	.0315	.2502	.7356	.3372	.7546	.1022
28.1856	.0315	.2497	.7372	.3380	.7540	.0912
32.2983	.0314	.2494	.7390	.3383	.7537	.0803
37.6090	.0314	.2490	.7417	.3375	.7543	.0695
44.2198	.0315	.2488	.7446	.3362	.7553	.0595
51.9287	.0316	.2486	.7467	.3355	.7557	.0510
60.9145	.0316	.2486	.7480	.3356	.7557	.0437
71.3941	.0316	.2486	.7490	.3358	.7555	.0375
83.6191	.0316	.2485	.7498	.3360	.7554	.0321
97.8823	.0316	.2485	.7505	.3362	.7553	.0275
114.5240	.0316	.2485	.7511	.3362	.7552	.0236
133.9415	.0316	.2485	.7517	.3363	.7552	.0202
156.5979	.0316	.2485	.7521	.3364	.7551	.0173
201.7512	.0316	.2486	.7527	.3365	.7551	.0135

MACH NO = 25.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	.0765	1.5198	-.1820	1.1325	1.0642
.7997	.0135	.0826	1.2661	-.1373	1.0781	1.0088
.9625	.0142	.8297	1.0905	-.0414	1.0302	.9519
1.1842	.0149	.7465	.9500	.0262	.9809	.8840
1.4321	.0154	.6902	.8593	.0825	.9403	.8187
1.7024	.0159	.6025	.8014	.1281	.9068	.7577
1.9895	.0162	.5439	.7657	.1637	.8809	.7021
2.2885	.0166	.4940	.7450	.1903	.8614	.6523
2.5957	.0171	.4523	.7357	.2095	.8482	.6079
2.9055	.0177	.4171	.7342	.2196	.8401	.5689
3.2139	.0184	.3884	.7383	.2249	.8363	.5348
3.5196	.0192	.3648	.7456	.2259	.8355	.5048
3.8179	.0202	.3454	.7548	.2239	.8370	.4785
4.6295	.0233	.3073	.7805	.2130	.8450	.4192
5.3433	.0261	.2863	.7944	.2077	.8488	.3781
6.0569	.0288	.2729	.8029	.2055	.8504	.3443
6.6999	.0313	.2655	.8086	.2043	.8513	.3186
7.3719	.0335	.2611	.8095	.2075	.8490	.2956
8.0111	.0349	.2589	.8045	.2166	.8423	.2765
8.7151	.0355	.2577	.7933	.2326	.8307	.2582
9.4179	.0354	.2572	.7789	.2522	.8164	.2422
10.2235	.0347	.2568	.7616	.2757	.7993	.2262
11.0434	.0338	.2563	.7456	.2976	.7833	.2119
11.9259	.0329	.2556	.7319	.3171	.7692	.1984
12.9438	.0321	.2546	.7222	.3323	.7581	.1848
13.9970	.0316	.2535	.7186	.3399	.7526	.1726
15.2400	.0314	.2523	.7197	.3419	.7511	.1601
16.5511	.0314	.2513	.7271	.3408	.7519	.1487
18.1299	.0315	.2503	.7272	.3388	.7533	.1370
19.9375	.0316	.2496	.7306	.3374	.7544	.1263
21.[illegible]	.0316	.2490	.7335	.3366	.7550	.1150
24.3[illegible]	.0316	.2486	.7354	.3359	.7548	.1045
27.5535	.0315	.2482	.7369	.3379	.7541	.0932
31.3509	.0314	.2478	.7385	.3385	.7536	.0826
35.2336	.0314	.2474	.7411	.3378	.7541	.0720
42.8[illegible]18	.0315	.2471	.7444	.3360	.7554	.0613
50.2870	.0316	.2469	.7463	.3350	.7562	.0526
59.4727	.0317	.2468	.7483	.3350	.7562	.0447
69.5826	.0317	.2468	.7491	.3354	.7559	.0384
82.1519	.0317	.2468	.7499	.3357	.7556	.0327
95.9932	.0317	.2468	.7505	.3360	.7554	.0281
113.2051	.0317	.2468	.7512	.3361	.7553	.0239
132.1603	.0317	.2468	.7517	.3362	.7552	.0205
155.7324	.0317	.2468	.7521	.3363	.7552	.0174
200.[illegible]466	.0317	.2468	.7527	.3364	.7551	.0136

MACH NO = 30.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 1.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0127	.2759	1.5198	-.1820	1.1325	1.0642
.7994	.0135	.2820	1.2665	-.1075	1.0782	1.0089
.9619	.0142	.2893	1.0910	-.0416	1.0303	.9521
1.1830	.0149	.7462	.9504	.0260	.9811	.8843
1.4304	.0154	.6700	.8596	.0823	.9401	.8191
1.7051	.0158	.6023	.8015	.1279	.9069	.7582
1.9864	.0162	.5437	.7657	.1635	.8810	.7026
2.2845	.0166	.4938	.7448	.1903	.8615	.6529
2.5907	.0170	.4518	.7355	.2085	.8482	.6086
2.8092	.0176	.4169	.7339	.2197	.8401	.5697
3.2063	.0183	.3881	.7379	.2250	.8362	.5356
3.5095	.0191	.3645	.7452	.2261	.8354	.5057
3.8072	.0201	.3450	.7545	.2240	.8369	.4794
4.6136	.0232	.3069	.7804	.2129	.8450	.4203
5.3642	.0262	.2848	.7950	.2072	.8491	.3770
6.0679	.0289	.2717	.8037	.2048	.8509	.3438
6.7012	.0314	.2644	.8095	.2034	.8520	.3185
7.3623	.0336	.2601	.8135	.2064	.8497	.2959
8.0299	.0350	.2579	.8050	.2161	.8427	.2760
8.7221	.0356	.2568	.7937	.2322	.8310	.2581
9.4126	.0355	.2564	.7792	.2519	.8167	.2424
10.2026	.0348	.2560	.7618	.2753	.7996	.2266
11.0564	.0338	.2555	.7448	.2986	.7826	.2117
11.9213	.0329	.2548	.7311	.3180	.7685	.1984
12.9171	.0321	.2538	.7215	.3330	.7575	.1851
14.0130	.0316	.2526	.7180	.3407	.7520	.1724
15.2310	.0314	.2514	.7194	.3423	.7509	.1602
16.5120	.0314	.2504	.7230	.3408	.7519	.1490
18.0483	.0315	.2495	.7272	.3387	.7534	.1376
19.8134	.0316	.2487	.7308	.3371	.7546	.1264
21.8836	.0316	.2481	.7336	.3364	.7551	.1154
24.2179	.0316	.2477	.7354	.3368	.7548	.1051
27.2791	.0315	.2477	.7367	.3378	.7541	.0941
31.1852	.0315	.2469	.7384	.3386	.7536	.0830
35.9394	.0315	.2465	.7409	.3378	.7541	.0726
42.4611	.0316	.2462	.7444	.3359	.7555	.0619
50.2264	.0317	.2460	.7472	.3345	.7565	.0527
59.3361	.0318	.2459	.7486	.3345	.7565	.0448
69.3519	.0318	.2459	.7494	.3350	.7561	.0385
81.7912	.0318	.2458	.7500	.3355	.7558	.0328
96.4013	.0318	.2458	.7507	.3358	.7555	.0279
113.5617	.0318	.2459	.7513	.3360	.7554	.0238
132.4409	.0318	.2459	.7517	.3362	.7553	.0204
155.8942	.0318	.2459	.7522	.3363	.7552	.0174
201.8449	.0318	.2459	.7528	.3364	.7551	.0135

MACH NO = 3.50 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9046	.0434	.9823	1.1055	-.0479	1.0084	1.0046
1.1272	.0478	.9508	.8989	.0561	.9902	.9853
1.4123	.0528	.9130	.7515	.1687	.9705	.9616
1.8745	.0598	.8579	.6312	.3199	.9440	.9256
2.4748	.0665	.7928	.5578	.4830	.9155	.8827
3.2328	.0719	.7228	.5124	.6573	.8850	.8339
4.1654	.0762	.6505	.4864	.8352	.8539	.7808
5.2880	.0794	.5795	.4753	1.0060	.8240	.7252
6.6153	.0818	.5126	.4736	1.1645	.7962	.6689
8.1627	.0834	.4518	.4785	1.3053	.7716	.6133
9.9462	.0846	.3982	.4883	1.4243	.7508	.5598
11.9547	.0854	.3607	.4983	1.5040	.7368	.5189
13.8085	.0862	.3203	.5120	1.5859	.7225	.4707
15.8191	.0867	.2929	.5234	1.6387	.7133	.4347
17.4555	.0870	.2748	.5320	1.6718	.7075	.4093
19.8188	.0875	.2539	.5432	1.7081	.7011	.3773
21.7344	.0878	.2402	.5514	1.7302	.6973	.3549
24.4908	.0882	.2245	.5619	1.7538	.6931	.3269
26.7179	.0886	.2143	.5694	1.7678	.6907	.3073
29.9134	.0891	.2026	.5789	1.7823	.6881	.2830
32.4889	.0895	.1950	.5857	1.7906	.6867	.2660
35.2260	.0899	.1884	.5920	1.7973	.6855	.2501
39.1413	.0905	.1807	.5999	1.8041	.6843	.2304
42.2881	.0909	.1758	.6053	1.8080	.6836	.2166
46.7818	.0915	.1702	.6119	1.8122	.6829	.1996
50.3882	.0919	.1665	.6165	1.8147	.6825	.1878
55.5312	.0925	.1624	.6220	1.8176	.6820	.1732
59.6536	.0929	.1597	.6257	1.8194	.6816	.1630
65.5261	.0935	.1566	.6303	1.8218	.6812	.1504
70.2286	.0939	.1546	.6334	1.8234	.6809	.1416
76.9214	.0944	.1523	.6371	1.8255	.6806	.1308
82.2765	.0947	.1509	.6396	1.8271	.6803	.1232
87.9359	.0951	.1496	.6419	1.8287	.6800	.1161
95.9829	.0955	.1481	.6447	1.8309	.6796	.1074
102.4160	.0958	.1471	.6466	1.8325	.6794	.1012
111.5589	.0962	.1460	.6488	1.8347	.6790	.0937
118.8652	.0965	.1453	.6504	1.8364	.6787	.0884
129.2458	.0968	.1444	.6522	1.8386	.6783	.0818
137.5393	.0971	.1439	.6535	1.8403	.6780	.0772
149.3205	.0974	.1433	.6550	1.8426	.6776	.0715
158.7323	.0976	.1429	.6560	1.8442	.6773	.0676
172.1016	.0978	.1424	.6573	1.8464	.6769	.0626
182.7825	.0980	.1421	.6581	1.8480	.6766	.0591
194.0590	.0981	.1418	.6589	1.8496	.6764	.0559
201.9243	.0982	.1417	.6594	1.8506	.6762	.0538

MACH NO = 5.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9119	.0426	.9346	1.0966	-.0442	1.0077	1.0039
1.1565	.0465	.9007	.8775	.0696	.9878	.9828
1.3959	.0498	.8694	.7530	.1660	.9710	.9630
1.9268	.0565	.8059	.6101	.3463	.9394	.9217
2.4885	.0595	.7464	.5383	.5055	.9114	.8818
3.1911	.0624	.6812	.4910	.6793	.8811	.8365
4.2940	.0647	.5958	.4577	.9011	.8423	.7740
5.3605	.0658	.5297	.4469	1.0701	.8127	.7219
6.5954	.0665	.4656	.4469	1.2214	.7863	.6696
8.3605	.0673	.3949	.4585	1.3755	.7593	.6062
10.0045	.0682	.3457	.4738	1.4691	.7429	.5582
11.8055	.0693	.3031	.4920	1.5383	.7308	.5130
14.3099	.0710	.2586	.5162	1.5954	.7207	.4612
16.5201	.0726	.2292	.5352	1.6258	.7155	.4234
18.3010	.0740	.2104	.5486	1.6409	.7129	.3972
20.8436	.0759	.1892	.5649	1.6551	.7104	.3650
23.5830	.0779	.1712	.5792	1.6653	.7086	.3356
26.5252	.0799	.1554	.5915	1.6738	.7071	.3089
28.8697	.0813	.1470	.5994	1.6799	.7061	.2905
32.1814	.0831	.1363	.6085	1.6881	.7046	.2680
35.7162	.0848	.1275	.6160	1.6957	.7031	.2475
39.4809	.0864	.1201	.6224	1.7057	.7015	.2288
42.4606	.0875	.1154	.6264	1.7126	.7003	.2159
46.6490	.0888	.1102	.6311	1.7219	.6987	.2001
51.0928	.0900	.1055	.6350	1.7311	.6971	.1857
55.8022	.0912	.1017	.6384	1.7401	.6955	.1725
59.5151	.0919	.0992	.6406	1.7467	.6944	.1633
64.7162	.0929	.0964	.6432	1.7552	.6929	.1520
70.2151	.0937	.0940	.6454	1.7633	.6915	.1417
76.0255	.0945	.0920	.6474	1.7710	.6901	.1321
80.5966	.0950	.0907	.6488	1.7764	.6892	.1255
86.9890	.0957	.0892	.6504	1.7834	.6879	.1173
93.7389	.0963	.0879	.6518	1.7899	.6868	.1097
100.8677	.0968	.0867	.6531	1.7961	.6857	.1027
108.4004	.0973	.0858	.6542	1.8019	.6847	.0962
114.3320	.0976	.0852	.6550	1.8060	.6840	.0916
122.6423	.0981	.0844	.6560	1.8113	.6831	.0859
131.4438	.0984	.0838	.6569	1.8162	.6822	.0805
140.7740	.0987	.0832	.6577	1.8210	.6814	.0756
148.1429	.0990	.0828	.6582	1.8244	.6808	.0721
158.4969	.0992	.0824	.6589	1.8287	.6800	.0676
169.4954	.0995	.0820	.6595	1.8327	.6793	.0635
181.1842	.0997	.0817	.6601	1.8365	.6787	.0596
190.4321	.0998	.0815	.6605	1.8392	.6782	.0569
200.1157	.1000	.0813	.6609	1.8417	.6777	.0543

MACH NO = 10.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9178	.0418	.9981	1.0955	-.0437	1.0077	1.0038
1.1618	.0451	.9642	.8726	.0727	.9873	.9823
1.5147	.0486	.8194	.7039	.2138	.9626	.9534
1.9616	.0518	.7674	.5926	.3673	.9357	.9192
2.6780	.0547	.6933	.5029	.5785	.8988	.8691
3.3865	.0551	.6310	.4566	.7588	.8672	.8247
4.4543	.0547	.5514	.4201	.9889	.8270	.7657
5.4587	.0536	.4897	.4039	1.1672	.7958	.7174
6.9016	.0518	.4183	.3975	1.3677	.7607	.6578
8.2017	.0505	.3670	.4014	1.5025	.7371	.6120
10.0006	.0490	.3110	.4146	1.6341	.7141	.5583
11.5720	.0481	.2726	.4301	1.7097	.7008	.5185
13.6939	.0475	.2318	.4533	1.7704	.6902	.4730
15.9849	.0473	.1981	.4790	1.7999	.6852	.4320
18.4442	.0478	.1704	.5058	1.8015	.6848	.3953
21.0770	.0488	.1475	.5325	1.7848	.6877	.3623
23.8949	.0503	.1286	.5593	1.7549	.6929	.3326
26.9152	.0524	.1129	.5825	1.7176	.6995	.3057
30.1581	.0550	.0999	.6045	1.6776	.7065	.2813
33.6433	.0581	.0890	.6239	1.6393	.7132	.2591
37.3853	.0615	.0800	.6403	1.6059	.7190	.2388
40.5708	.0644	.0739	.6512	1.5842	.7228	.2239
44.7902	.0682	.0674	.6622	1.5645	.7263	.2068
49.2673	.0719	.0621	.6705	1.5529	.7283	.1913
53.9911	.0755	.0578	.6765	1.5486	.7290	.1773
58.8784	.0789	.0542	.6805	1.5504	.7287	.1648
63.8157	.0820	.0514	.6830	1.5563	.7277	.1539
68.8287	.0847	.0491	.6844	1.5653	.7261	.1441
73.9672	.0872	.0471	.6851	1.5764	.7242	.1354
79.2927	.0894	.0455	.6852	1.5892	.7219	.1273
84.8573	.0914	.0441	.6848	1.6035	.7194	.1199
90.6977	.0932	.0430	.6840	1.6192	.7167	.1130
96.8413	.0947	.0419	.6828	1.6360	.7137	.1065
103.3134	.0960	.0410	.6814	1.6534	.7107	.1005
110.1411	.0971	.0402	.6798	1.6708	.7076	.0948
117.3525	.0981	.0395	.6783	1.6879	.7047	.0894
124.9756	.0989	.0389	.6768	1.7042	.7018	.0844
133.0393	.0997	.0384	.6754	1.7199	.6991	.0796
139.8241	.1002	.0380	.6743	1.7318	.6970	.0760
148.7497	.1007	.0376	.6730	1.7460	.6945	.0718
158.1942	.1011	.0372	.6718	1.7594	.6921	.0678
168.1863	.1015	.0369	.6706	1.7719	.6899	.0640
178.7559	.1018	.0366	.6696	1.7835	.6879	.0604
189.9355	.1020	.0364	.6687	1.7942	.6861	.0570
201.7636	.1022	.0362	.6680	1.8039	.6844	.0539

MACH NO = 15.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

		INVISCID	AERODYNAMIC	COEFFICIENTS		
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0417	.8912	1.0955	-.0437	1.0077	1.0038
1.1556	.0447	.8582	.8764	.0702	.9877	.9829
1.5026	.0480	.8142	.7069	.2102	.9632	.9544
2.0687	.0515	.7491	.5718	.4037	.9294	.9113
2.6415	.0531	.6909	.5019	.5733	.8997	.8715
3.5214	.0535	.6137	.4441	.7994	.8601	.8167
4.5938	.0525	.5360	.4082	1.0312	.8196	.7586
5.8606	.0505	.4620	.3885	1.2539	.7806	.6998
7.0095	.0487	.4076	.3826	1.4146	.7525	.6538
8.6037	.0464	.3472	.3852	1.5847	.7227	.5991
10.3576	.0445	.2955	.3960	1.7163	.6997	.5487
11.8634	.0432	.2603	.4088	1.7943	.6860	.5117
13.8595	.0419	.2229	.4283	1.8610	.6744	.4697
16.4015	.0410	.1866	.4546	1.9024	.6671	.4253
19.5418	.0406	.1536	.4869	1.9092	.6659	.3808
22.3634	.0409	.1316	.5145	1.8895	.6694	.3481
25.2915	.0417	.1139	.5413	1.8540	.6756	.3196
28.3168	.0429	.0997	.5665	1.8087	.6835	.2947
31.9607	.0450	.0864	.5935	1.7496	.6939	.2694
35.1754	.0472	.0773	.6145	1.6979	.7029	.2504
38.4675	.0498	.0698	.6331	1.6482	.7116	.2335
41.8280	.0527	.0636	.6493	1.6030	.7195	.2185
45.8204	.0564	.0578	.6651	1.5578	.7274	.2030
49.2913	.0598	.0537	.6761	1.5265	.7329	.1912
52.8006	.0632	.0502	.6850	1.5020	.7372	.1806
56.3542	.0666	.0474	.6921	1.4839	.7403	.1710
60.5952	.0705	.0445	.6983	1.4699	.7428	.1608
64.3765	.0739	.0425	.7023	1.4634	.7439	.1527
68.3701	.0772	.0406	.7051	1.4616	.7443	.1450
72.6511	.0805	.0390	.7067	1.4648	.7437	.1375
78.1062	.0841	.0372	.7072	1.4753	.7419	.1290
83.2554	.0870	.0359	.7064	1.4904	.7392	.1219
88.8986	.0896	.0347	.7046	1.5103	.7357	.1150
95.0100	.0920	.0336	.7021	1.5335	.7317	.1084
102.6045	.0943	.0325	.6988	1.5621	.7267	.1011
109.5554	.0960	.0316	.6957	1.5871	.7223	.0952
116.9457	.0974	.0309	.6926	1.6123	.7179	.0897
124.7955	.0986	.0303	.6895	1.6371	.7135	.0845
134.5566	.0997	.0296	.6860	1.6650	.7087	.0788
143.4645	.1005	.0291	.6832	1.6875	.7047	.0743
152.9016	.1011	.0287	.6806	1.7084	.7011	.0700
162.8991	.1016	.0283	.6783	1.7277	.6977	.0659
173.4856	.1020	.0280	.6762	1.7453	.6946	.0621
186.6292	.1024	.0277	.6740	1.7638	.6914	.0580
200.6801	.1027	.0274	.6722	1.7802	.6885	.0541

MACH NO = 20.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0416	.8886	1.0955	-.0437	1.0077	1.0038
1.1533	.0446	.8559	.8779	.0692	.9879	.9831
1.4981	.0478	.8122	.7081	.2088	.9635	.9547
2.0604	.0511	.7476	.5722	.4019	.9297	.9119
2.6278	.0526	.6898	.5017	.5712	.9001	.8724
3.4982	.0530	.6132	.4430	.7970	.8605	.8181
4.5567	.0518	.5361	.4058	1.0295	.8199	.7605
5.8042	.0497	.4627	.3845	1.2542	.7805	.7022
6.9326	.0477	.4088	.3771	1.4178	.7519	.6567
8.4933	.0453	.3488	.3775	1.5930	.7213	.6026
10.2029	.0431	.2975	.3861	1.7313	.6971	.5528
11.6643	.0416	.2624	.3971	1.8155	.6823	.5163
13.5917	.0401	.2253	.4143	1.8906	.6692	.4750
16.8663	.0385	.1789	.4462	1.9525	.6584	.4181
19.8801	.0378	.1484	.4758	1.9619	.6567	.3766
22.9931	.0378	.1251	.5052	1.9429	.6600	.3415
26.6305	.0383	.1048	.5371	1.8989	.6677	.3081
29.8523	.0393	.0912	.5630	1.8489	.6765	.2834
33.5493	.0410	.0791	.5901	1.7853	.6876	.2596
36.7783	.0430	.0708	.6115	1.7282	.6976	.2419
40.4443	.0456	.0633	.6332	1.6650	.7087	.2245
43.6196	.0482	.0580	.6498	1.6139	.7176	.2113
46.7573	.0511	.0537	.6641	1.5682	.7256	.1997
50.2946	.0545	.0497	.6779	1.5234	.7334	.1881
53.3547	.0577	.0468	.6879	1.4909	.7391	.1791
56.8402	.0613	.0440	.6973	1.4609	.7444	.1698
59.9222	.0646	.0419	.7039	1.4404	.7480	.1624
63.5501	.0684	.0399	.7099	1.4232	.7510	.1544
66.8822	.0718	.0383	.7139	1.4137	.7526	.1478
70.4252	.0751	.0368	.7166	1.4099	.7533	.1413
74.8034	.0788	.0353	.7181	1.4130	.7527	.1340
78.9390	.0819	.0340	.7181	1.4224	.7511	.1278
84.1080	.0862	.0327	.7167	1.4400	.7480	.1208
89.1263	.0878	.0317	.7144	1.4605	.7444	.1148
95.5837	.0906	.0305	.7110	1.4886	.7395	.1078
102.0184	.0928	.0296	.7074	1.5167	.7346	.1016
109.2492	.0948	.0287	.7033	1.5475	.7292	.0955
118.3810	.0967	.0278	.6984	1.5837	.7229	.0887
126.9353	.0981	.0270	.6942	1.6143	.7175	.0832
137.3775	.0993	.0264	.6899	1.6469	.7118	.0773
147.1249	.1002	.0258	.6864	1.6731	.7072	.0725
159.0215	.1010	.0253	.6829	1.7004	.7025	.0674
170.1246	.1016	.0249	.6802	1.7219	.6987	.0633
181.9236	.1021	.0246	.6778	1.7412	.6953	.0594
200.0837	.1026	.0242	.6749	1.7655	.6911	.0543

MACH NO = 25.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0416	.8875	1.0955	-.0437	1.0077	1.0038
1.1523	.0445	.8551	.8786	.0698	.9880	.9831
1.4961	.0477	.8115	.7086	.2081	.9636	.9549
2.0564	.0510	.7471	.5724	.4011	.9298	.9122
2.7800	.0526	.6745	.4875	.6143	.8925	.8624
3.6825	.0526	.5977	.4335	.8422	.8526	.8074
4.5391	.0515	.5364	.4048	1.0285	.8200	.7614
5.7774	.0493	.4632	.3827	1.2543	.7805	.7033
7.1925	.0468	.3968	.3736	1.4575	.7450	.6470
8.7678	.0443	.3388	.3748	1.6282	.7151	.5940
10.4829	.0421	.2893	.3835	1.7623	.6916	.5454
11.9414	.0406	.2556	.3943	1.8440	.6773	.5099
13.8556	.0391	.2200	.4110	1.9171	.6645	.4698
17.5010	.0372	.1707	.4456	1.9823	.6531	.4086
20.8909	.0364	.1391	.4778	1.9879	.6522	.3644
24.3585	.0363	.1157	.5090	1.9621	.6567	.3281
28.2877	.0368	.0963	.5418	1.9139	.6656	.2949
31.7586	.0379	.0834	.5685	1.8544	.6755	.2706
35.1760	.0394	.0735	.5928	1.7933	.6862	.2504
38.9305	.0416	.0649	.6172	1.7238	.6984	.2314
42.1742	.0439	.0590	.6364	1.6646	.7087	.2171
45.7114	.0468	.0537	.6552	1.6034	.7194	.2034
48.7580	.0497	.0500	.6695	1.5550	.7279	.1930
51.7235	.0527	.0469	.6817	1.5129	.7353	.1838
54.9900	.0562	.0440	.6932	1.4728	.7423	.1746
57.8704	.0594	.0419	.7018	1.4433	.7475	.1672
60.7767	.0627	.0400	.7089	1.4193	.7517	.1604
64.1436	.0665	.0382	.7153	1.3988	.7552	.1532
67.2871	.0699	.0367	.7196	1.3868	.7573	.1470
70.6372	.0733	.0354	.7225	1.3812	.7583	.1409
74.6659	.0770	.0340	.7240	1.3831	.7580	.1343
78.5168	.0800	.0328	.7240	1.3918	.7565	.1284
83.2639	.0833	.0316	.7225	1.4086	.7535	.1219
87.9692	.0860	.0305	.7202	1.4289	.7500	.1161
93.2667	.0886	.0295	.7171	1.4538	.7456	.1102
99.8899	.0912	.0285	.7128	1.4854	.7401	.1036
106.4198	.0932	.0276	.7086	1.5159	.7347	.0978
113.7623	.0950	.0268	.7041	1.5484	.7291	.0920
123.3151	.0968	.0259	.6988	1.5867	.7224	.0854
132.9705	.0982	.0252	.6941	1.6203	.7165	.0797
143.2801	.0993	.0246	.6900	1.6511	.7111	.0743
155.7060	.1003	.0240	.6859	1.682.	.7057	.0688
167.5467	.1011	.0235	.6827	1.7068	.7013	.0642
181.8401	.1017	.0231	.6796	1.7315	.6970	.0594
200.8351	.1024	.0227	.6764	1.7577	.6924	.0541

MACH NO = 30.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9128	.0416	.9869	1.0955	-.0437	1.0077	1.0038
1.1517	.0445	.8545	.8789	.0685	.9880	.9832
1.4949	.0477	.8110	.7089	.2078	.9636	.9550
2.0541	.0509	.7467	.5725	.4006	.9299	.9124
2.7759	.0525	.6742	.4875	.6137	.8926	.8627
3.6758	.0524	.5977	.4332	.8415	.8528	.8078
4.5294	.0513	.5364	.4042	1.0280	.8201	.7618
5.7626	.0491	.4635	.3817	1.2542	.7805	.7040
7.1708	.0465	.3972	.3721	1.4584	.7448	.6478
8.7368	.0440	.3392	.3727	1.6305	.7147	.5950
10.4397	.0418	.2899	.3808	1.7665	.6909	.5465
11.8862	.0402	.2562	.3911	1.8498	.6763	.5112
13.7821	.0386	.2207	.4071	1.9254	.6631	.4712
17.3835	.0366	.1715	.4405	1.9952	.6509	.4103
21.1425	.0356	.1366	.4756	2.0041	.6493	.3615
25.3957	.0354	.1094	.5130	1.9707	.6552	.3187
29.2102	.0360	.0920	.5438	1.9190	.6642	.2880
32.9515	.0371	.0791	.5719	1.8567	.6751	.2632
36.5804	.0387	.0694	.5971	1.7900	.6868	.2429
40.0746	.0407	.0620	.6196	1.7233	.6985	.2261
43.4252	.0431	.0563	.6395	1.6598	.7096	.2121
46.6338	.0458	.0518	.6568	1.6016	.7198	.2001
49.7116	.0487	.0482	.6717	1.5495	.7289	.1899
52.6804	.0517	.0452	.6845	1.5041	.7368	.1810
55.5739	.0549	.0428	.6953	1.4651	.7436	.1731
58.4369	.0582	.0407	.7045	1.4323	.7494	.1659
61.6474	.0620	.0388	.7128	1.4029	.7545	.1585
64.6283	.0655	.0372	.7188	1.3830	.7580	.1522
67.7629	.0690	.0358	.7233	1.3697	.7603	.1461
71.1055	.0725	.0344	.7262	1.3639	.7614	.1401
74.6424	.0759	.0332	.7275	1.3656	.7611	.1343
78.4424	.0791	.0321	.7274	1.3743	.7595	.1285
82.6036	.0820	.0310	.7260	1.3895	.7569	.1228
87.2493	.0848	.0300	.7236	1.4104	.7532	.1170
92.4576	.0875	.0290	.7203	1.4360	.7487	.1110
98.0950	.0898	.0280	.7164	1.4643	.7438	.1053
104.3011	.0920	.0272	.7121	1.4950	.7384	.0996
112.1268	.0940	.0262	.7069	1.5317	.7320	.0932
120.3228	.0957	.0254	.7020	1.5667	.7259	.0874
129.9976	.0973	.0246	.6969	1.6029	.7195	.0814
141.0236	.0986	.0239	.6921	1.6379	.7134	.0754
152.8527	.0997	.0233	.6879	1.6690	.7080	.0700
165.5579	.1006	.0228	.6843	1.6967	.7031	.0649
179.2185	.1014	.0224	.6812	1.7213	.6988	.0603
200.8037	.1022	.0218	.6774	1.7523	.6934	.0541

MACH NO = 3.50 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0432	.9839	1.1167	-.0526	1.0110	1.0055
1.1150	.0474	.9469	.9099	.0494	.9896	.9827
1.3949	.0523	.9034	.7633	.1577	.9669	.9551
1.7459	.0576	.8537	.6634	.2711	.9430	.9226
2.3003	.0638	.7845	.5836	.4193	.9119	.8755
2.9994	.0690	.7102	.5347	.5740	.8793	.8226
3.6729	.0722	.6499	.5112	.6977	.8533	.7773
4.6680	.0756	.5765	.4976	.8430	.8228	.7189
5.8444	.0782	.5082	.4951	.9741	.7952	.6602
6.9237	.0798	.4586	.4983	1.0667	.7758	.6142
8.4555	.0815	.4036	.5074	1.1641	.7553	.5589
9.8347	.0827	.3655	.5171	1.2278	.7419	.5170
11.7621	.0841	.3249	.5305	1.2923	.7284	.4680
13.4758	.0851	.2975	.5415	1.3335	.7197	.4316
15.3509	.0861	.2742	.5522	1.3672	.7126	.3978
17.3960	.0869	.2544	.5625	1.3945	.7069	.3665
19.6203	.0877	.2377	.5722	1.4167	.7022	.3375
21.4122	.0883	.2270	.5790	1.4305	.6993	.3174
23.9748	.0891	.2147	.5875	1.4456	.6961	.2924
26.7459	.0898	.2044	.5954	1.4577	.6936	.2694
29.7377	.0905	.1957	.6027	1.4674	.6915	.2484
32.9631	.0911	.1885	.6093	1.4753	.6899	.2291
36.4360	.0918	.1824	.6153	1.4818	.6885	.2114
39.2121	.0922	.1785	.6194	1.4860	.6876	.1991
43.1548	.0928	.1740	.6244	1.4908	.6866	.1840
47.3885	.0934	.1703	.6290	1.4949	.6858	.1700
51.9314	.0940	.1671	.6330	1.4985	.6850	.1573
56.8027	.0945	.1644	.6367	1.5018	.6843	.1455
62.0232	.0949	.1622	.6400	1.5048	.6837	.1348
67.6154	.0954	.1603	.6429	1.5075	.6831	.1249
72.0679	.0957	.1591	.6450	1.5095	.6827	.1180
78.3699	.0961	.1577	.6474	1.5120	.6822	.1094
85.1149	.0965	.1565	.6496	1.5144	.6817	.1016
92.3334	.0968	.1554	.6516	1.5166	.6812	.0943
100.0584	.0971	.1546	.6534	1.5188	.6807	.0876
108.3260	.0974	.1539	.6550	1.5210	.6803	.0814
114.9063	.0976	.1534	.6561	1.5225	.6800	.0771
124.2203	.0979	.1528	.6574	1.5246	.6795	.0717
134.1936	.0981	.1524	.6586	1.5265	.6791	.0666
144.8760	.0983	.1520	.6597	1.5285	.6787	.0620
156.3210	.0984	.1516	.6607	1.5303	.6783	.0577
168.5866	.0986	.1513	.6616	1.5321	.6779	.0537
181.7349	.0987	.1511	.6624	1.5337	.6776	.0500
192.2155	.0988	.1509	.6630	1.5349	.6773	.0474
203.2602	.0989	.1508	.6635	1.5361	.6771	.0449

MACH NO = 5.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0423	.9371	1.1167	-.0526	1.0110	1.0055
1.1330	.0461	.8980	.8959	.0578	.9878	.9809
1.3646	.0492	.8624	.7705	.1500	.9685	.9580
1.7592	.0535	.8068	.6501	.2836	.9404	.9214
2.2648	.0573	.7435	.5708	.4269	.9103	.8784
2.8938	.0601	.6753	.5187	.5781	.8785	.8302
3.8644	.0623	.5883	.4822	.7657	.8390	.7654
4.7958	.0634	.5211	.4698	.9052	.8097	.7120
5.8726	.0641	.4583	.4686	1.0282	.7839	.6589
7.1011	.0649	.4015	.4760	1.1297	.7625	.6073
8.4889	.0658	.3511	.4896	1.2085	.7460	.5578
10.0332	.0670	.3076	.5066	1.2660	.7339	.5115
12.1736	.0689	.2627	.5298	1.3128	.7240	.4587
14.0542	.0708	.2333	.5482	1.3355	.7193	.4206
16.0849	.0728	.2089	.5652	1.3496	.7163	.3860
18.2672	.0749	.1888	.5804	1.3588	.7144	.3546
21.2111	.0776	.1685	.5966	1.3670	.7126	.3195
23.7418	.0797	.1555	.6073	1.3730	.7114	.2945
26.4320	.0817	.1447	.6161	1.3793	.7101	.2718
29.2851	.0836	.1359	.6234	1.3862	.7086	.2514
33.0865	.0858	.1269	.6306	1.3957	.7066	.2284
36.3211	.0873	.1210	.6353	1.4038	.7049	.2120
39.7330	.0887	.1161	.6392	1.4120	.7032	.1970
43.3276	.0899	.1121	.6424	1.4203	.7014	.1833
48.0867	.0913	.1079	.6457	1.4305	.6993	.1679
52.1151	.0923	.1051	.6480	1.4383	.6977	.1568
56.3485	.0932	.1027	.6499	1.4458	.6961	.1466
60.7964	.0940	.1008	.6515	1.4530	.6946	.1372
66.6757	.0949	.0987	.6533	1.4614	.6928	.1264
71.6524	.0956	.0973	.6546	1.4678	.6915	.1186
76.8913	.0961	.0961	.6557	1.4737	.6902	.1113
82.4131	.0967	.0951	.6567	1.4794	.6890	.1046
89.7487	.0972	.0940	.6578	1.4861	.6876	.0968
95.9947	.0976	.0932	.6586	1.4912	.6865	.0910
102.6048	.0980	.0926	.6593	1.4960	.6855	.0856
109.6060	.0983	.0920	.6599	1.5006	.6846	.0805
117.0261	.0986	.0915	.6605	1.5048	.6837	.0758
126.9332	.0989	.0910	.6612	1.5098	.6826	.0702
135.4001	.0991	.0906	.6617	1.5135	.6818	.0661
144.3812	.0993	.0903	.6622	1.5169	.6811	.0622
153.9087	.0995	.0900	.6627	1.5201	.6805	.0586
166.6383	.0996	.0897	.6632	1.5237	.6797	.0543
177.5226	.0998	.0895	.6637	1.5263	.6792	.0511
189.0717	.0999	.0893	.6641	1.5287	.6787	.0481
201.3268	.1000	.0892	.6645	1.5309	.6782	.0453

MACH NO = 10.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0416	.9018	1.1167	-.0526	1.0110	1.0055
1.1335	.0446	.8622	.8941	.0589	.9876	.9808
1.3791	.0472	.8250	.7602	.1582	.9667	.9566
1.8914	.0518	.7548	.6128	.3331	.9300	.9097
2.4053	.0526	.6931	.5386	.4811	.8989	.8671
3.1928	.0535	.6124	.4787	.6733	.8585	.8091
3.9453	.0532	.5490	.4491	.8264	.8263	.7604
5.0417	.0521	.4715	.4287	1.0070	.7883	.6991
6.0407	.0509	.4154	.4234	1.1343	.7616	.6513
7.4335	.0495	.3533	.4278	1.2647	.7341	.5946
8.6549	.0486	.3103	.4378	1.3441	.7175	.5525
9.9637	.0479	.2731	.4519	1.4021	.7053	.5134
11.7123	.0475	.2338	.4731	1.4476	.6957	.4692
13.9613	.0477	.1958	.5014	1.4701	.6910	.4223
15.9525	.0484	.1704	.5255	1.4687	.6913	.3880
18.4772	.0498	.1459	.5537	1.4507	.6951	.3518
21.1506	.0519	.1264	.5800	1.4220	.7011	.3202
23.9787	.0545	.1109	.6038	1.3888	.7081	.2923
26.4597	.0571	.1003	.6213	1.3611	.7139	.2716
29.5920	.0606	.0899	.6393	1.3308	.7202	.2493
32.8969	.0644	.0815	.6540	1.3060	.7255	.2295
36.3733	.0683	.0747	.6655	1.2878	.7293	.2117
40.0178	.0722	.0692	.6743	1.2764	.7317	.1958
43.1841	.0754	.0654	.6797	1.2717	.7327	.1838
47.1540	.0791	.0617	.6843	1.2709	.7328	.1708
51.3561	.0825	.0586	.6874	1.2747	.7321	.1588
55.8752	.0858	.0560	.6892	1.2824	.7304	.1476
59.8873	.0883	.0541	.6897	1.2914	.7285	.1390
64.9128	.0910	.0523	.6895	1.3048	.7257	.1295
70.2127	.0933	.0507	.6885	1.3207	.7224	.1208
75.8260	.0952	.0494	.6869	1.3383	.7187	.1127
80.7704	.0965	.0485	.6853	1.3537	.7154	.1065
87.0560	.0979	.0476	.6832	1.3722	.7116	.0995
93.7607	.0990	.0468	.6810	1.3904	.7077	.0930
100.9208	.0999	.0461	.6789	1.4080	.7040	.0869
107.2610	.1005	.0456	.6772	1.4221	.7011	.0821
115.3470	.1011	.0452	.6753	1.4381	.6977	.0768
123.9862	.1015	.0446	.6735	1.4528	.6946	.0718
133.2144	.1019	.0442	.6720	1.4653	.6918	.0671
141.3830	.1021	.0439	.6708	1.4765	.6896	.0635
151.7962	.1023	.0436	.6697	1.4875	.6873	.0593
162.9188	.1024	.0434	.6687	1.4972	.6853	.0555
174.7985	.1025	.0432	.6680	1.5057	.6835	.0519
185.3133	.1025	.0430	.6675	1.5120	.6822	.0491
201.0374	.1025	.0428	.6669	1.5198	.6805	.0454

MACH NO = 15.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0415	.8938	1.1167	-.0526	1.0110	1.0055
1.1281	.0443	.8562	.8977	.0566	.9881	.9814
1.4588	.0474	.8067	.7275	.1886	.9603	.9490
1.8706	.0499	.7509	.6146	.3286	.9309	.9116
2.5113	.0518	.6749	.5240	.5134	.8921	.8588
3.1351	.0521	.6115	.4764	.6673	.8597	.8131
4.0601	.0513	.5329	.4381	.8595	.8193	.7535
4.9162	.0500	.4732	.4199	1.0062	.7885	.7057
6.1248	.0481	.4050	.4101	1.1698	.7541	.6476
7.1936	.0465	.3567	.4106	1.2797	.7310	.6037
8.6424	.0448	.3045	.4191	1.3878	.7083	.5529
9.8799	.0437	.2689	.4303	1.4514	.6949	.5158
11.5100	.0427	.2314	.4480	1.5054	.6836	.4739
13.9189	.0419	.1898	.4765	1.5417	.6759	.4231
16.8109	.0420	.1544	.5105	1.5426	.6757	.3749
19.4302	.0427	.1311	.5394	1.5206	.6804	.3398
22.1080	.0440	.1132	.5664	1.4865	.6875	.3102
25.2161	.0461	.0975	.5943	1.4406	.6972	.2816
27.9614	.0484	.0869	.6158	1.3989	.7059	.2605
31.1142	.0515	.0774	.6372	1.3532	.7155	.2398
33.8784	.0546	.0708	.6532	1.3170	.7232	.2242
36.6442	.0579	.0655	.6666	1.2854	.7298	.2104
39.8098	.0618	.0606	.6792	1.2558	.7360	.1967
42.5957	.0653	.0571	.6881	1.2354	.7403	.1860
45.4209	.0688	.0542	.6952	1.2199	.7436	.1762
48.7474	.0729	.0514	.7015	1.2078	.7461	.1660
51.8004	.0764	.0492	.7055	1.2020	.7473	.1576
55.0460	.0799	.0474	.7083	1.2008	.7476	.1496
59.0708	.0838	.0454	.7098	1.2055	.7466	.1407
62.9408	.0870	.0439	.7098	1.2151	.7446	.1331
67.8234	.0903	.0424	.7083	1.2320	.7410	.1245
72.5955	.0930	.0411	.7059	1.2512	.7370	.1172
77.9489	.0953	.0400	.7028	1.2738	.7322	.1100
84.9063	.0976	.0388	.6985	1.3028	.7261	.1018
91.5552	.0992	.0379	.6945	1.3291	.7206	.0950
98.6815	.1004	.0372	.6905	1.3548	.7152	.0887
107.4399	.1015	.0364	.6862	1.3825	.7094	.0820
115.6719	.1022	.0359	.6828	1.4049	.7047	.0766
124.4687	.1027	.0354	.6798	1.4251	.7004	.0715
135.2601	.1032	.0349	.6768	1.4457	.6961	.0662
145.3932	.1034	.0346	.6746	1.4617	.6927	.0618
157.8227	.1036	.0343	.6725	1.4776	.6894	.0572
169.4972	.1037	.0341	.6710	1.4897	.6868	.0534
181.9733	.1038	.0339	.6698	1.5004	.6846	.0499
201.3011	.1038	.0336	.6684	1.5133	.6819	.0453

MACH NO = 20.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0414	.8912	1.1167	-.0526	1.0110	1.0055
1.1261	.0442	.8539	.8990	.0558	.9883	.9816
1.4543	.0472	.8048	.7288	.1872	.9606	.9494
1.8624	.0496	.7435	.6155	.3267	.9313	.9123
2.4969	.0514	.6740	.5240	.5111	.8926	.8600
3.2838	.0515	.5954	.4662	.7037	.8521	.8029
4.0271	.0507	.5331	.4362	.8576	.8197	.7555
5.0967	.0489	.4597	.4136	1.0406	.7813	.6963
6.3129	.0468	.3936	.4045	1.2018	.7474	.6394
7.3806	.0451	.3470	.4051	1.3099	.7247	.5966
8.8174	.0433	.2968	.4130	1.4165	.7022	.5473
10.0361	.0421	.2627	.4234	1.4798	.6889	.5114
11.6304	.0409	.2268	.4399	1.5344	.6774	.4711
14.6487	.0397	.1774	.4742	1.5784	.6682	.4098
17.4358	.0395	.1458	.5059	1.5762	.6687	.3659
20.6256	.0401	.1201	.5398	1.5468	.6748	.3259
23.8188	.0415	.1014	.5705	1.5027	.6841	.2938
26.6245	.0432	.0890	.5948	1.4583	.6935	.2704
29.7176	.0456	.0785	.6189	1.4072	.7042	.2485
32.7289	.0484	.0705	.6398	1.3582	.7145	.2304
35.3345	.0512	.0649	.6557	1.3182	.7229	.2167
38.1926	.0547	.0599	.6710	1.2784	.7313	.2035
40.6819	.0579	.0563	.6824	1.2480	.7377	.1932
43.4515	.0616	.0530	.6931	1.2195	.7436	.1829
46.2289	.0655	.0503	.7018	1.1968	.7484	.1736
48.7468	.0690	.0482	.7080	1.1814	.7517	.1660
51.6903	.0729	.0461	.7134	1.1695	.7542	.1579
54.8174	.0768	.0443	.7171	1.1638	.7554	.1501
57.8098	.0803	.0428	.7190	1.1643	.7553	.1433
61.4370	.0839	.0413	.7195	1.1711	.7538	.1359
65.4171	.0873	.0400	.7185	1.1842	.7511	.1286
69.3462	.0901	.0388	.7164	1.2007	.7476	.1221
74.3372	.0930	.0376	.7130	1.2240	.7427	.1148
79.3116	.0953	.0366	.7093	1.2481	.7376	.1083
85.5774	.0974	.0355	.7045	1.2779	.7314	.1011
92.7881	.0993	.0346	.6992	1.3098	.7247	.0939
100.2353	.1006	.0338	.6944	1.3392	.7185	.0874
109.5414	.1018	.0330	.6893	1.3707	.7119	.0806
119.5990	.1027	.0323	.6849	1.3987	.7060	.0742
129.2174	.1033	.0318	.6815	1.4209	.7013	.0691
140.8642	.1038	.0314	.6783	1.4429	.6967	.0637
153.4613	.1041	.0310	.6755	1.4620	.6927	.0587
165.5251	.1043	.0308	.6735	1.4770	.6895	.0547
180.1538	.1044	.0305	.6716	1.4916	.6865	.0504
201.5615	.1044	.0302	.6696	1.5078	.6831	.0453

MACH NO = 25.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0414	.0902	1.1167	-.0526	1.0110	1.0055
1.1252	.0441	.0831	.8996	.0554	.9883	.9817
1.4522	.0471	.0842	.7294	.1866	.9608	.9496
1.8586	.0495	.0790	.6159	.3258	.9315	.9126
2.4902	.0512	.0738	.5240	.5100	.8928	.8605
3.2728	.0513	.0654	.4658	.7025	.8523	.8036
4.0115	.0504	.0534	.4354	.8566	.8199	.7564
5.0736	.0486	.0401	.4120	1.0404	.7813	.6975
6.2795	.0464	.0343	.4021	1.2030	.7471	.6409
7.3367	.0447	.0278	.4019	1.3126	.7241	.5983
8.7570	.0427	.0278	.4088	1.4216	.7012	.5492
9.9596	.0414	.0237	.4185	1.4871	.6874	.5136
11.5295	.0401	.0279	.4340	1.5447	.6753	.4734
14.8273	.0386	.0140	.4706	1.5962	.6645	.4067
17.8963	.0383	.0105	.5048	1.5933	.6651	.3595
21.3289	.0388	.0143	.5403	1.5601	.6720	.3183
24.7109	.0401	.0160	.5720	1.5121	.6822	.2859
27.6655	.0418	.0139	.5970	1.4637	.6923	.2626
30.8209	.0442	.0140	.6214	1.4092	.7038	.2416
33.5385	.0468	.0173	.6405	1.3622	.7137	.2260
36.4292	.0499	.0215	.6588	1.3142	.7238	.2115
39.1936	.0533	.0269	.6743	1.2717	.7327	.1992
41.6172	.0565	.0337	.6860	1.2386	.7396	.1896
44.2541	.0603	.0407	.6970	1.2076	.7462	.1801
46.6209	.0637	.0485	.7052	1.1847	.7510	.1724
49.2936	.0676	.0463	.7126	1.1649	.7551	.1644
52.0711	.0716	.0444	.7182	1.1513	.7580	.1569
54.7173	.0751	.0429	.7217	1.1446	.7594	.1503
57.8607	.0790	.0414	.7240	1.1437	.7596	.1432
60.8874	.0823	.0401	.7245	1.1489	.7585	.1370
64.5562	.0858	.0388	.7236	1.1607	.7560	.1301
68.6611	.0889	.0376	.7214	1.1785	.7523	.1232
72.7926	.0916	.0365	.7183	1.1990	.7480	.1169
77.8034	.0941	.0354	.7141	1.2250	.7425	.1102
82.8200	.0961	.0345	.7099	1.2509	.7371	.1041
89.1670	.0981	.0335	.7047	1.2819	.7305	.0973
96.6175	.0997	.0326	.6992	1.3146	.7237	.0905
104.5775	.1010	.0318	.6941	1.3449	.7173	.0841
114.7268	.1022	.0310	.6889	1.3770	.7105	.0772
124.7238	.1031	.0305	.6849	1.4029	.7051	.0714
136.7372	.1038	.0299	.6810	1.4281	.6998	.0655
149.7964	.1042	.0295	.6777	1.4502	.6952	.0601
162.5296	.1045	.0292	.6752	1.4676	.6915	.0556
177.8612	.1046	.0289	.6729	1.4845	.6879	.0510
201.6346	.1047	.0286	.6704	1.5038	.6839	.0453

MACH NO = 30.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8955	.0414	.8895	1.1167	-.0526	1.0110	1.0055
1.1246	.0441	.8525	.8999	.0552	.9884	.9817
1.4509	.0470	.8037	.7298	.1862	.9609	.9498
1.8564	.0494	.7487	.6161	.3252	.9316	.9128
2.4863	.0511	.6737	.5240	.5094	.8929	.8608
3.2666	.0512	.5954	.4656	.7018	.8525	.8040
4.0027	.0503	.5335	.4349	.8561	.8200	.7569
5.0605	.0484	.4604	.4112	1.0402	.7813	.6982
6.2607	.0462	.3946	.4008	1.2036	.7470	.6417
7.3121	.0444	.3483	.4002	1.3140	.7238	.5992
8.7233	.0424	.2983	.4066	1.4243	.7006	.5503
9.9169	.0411	.2643	.4158	1.4911	.6866	.5147
11.4733	.0397	.2285	.4308	1.5503	.6741	.4748
14.7358	.0381	.1747	.4664	1.6053	.6626	.4083
18.4374	.0376	.1351	.5070	1.6013	.6634	.3523
21.7993	.0381	.1108	.5411	1.5672	.6706	.3133
25.0798	.0393	.0936	.5713	1.5201	.6805	.2828
28.5353	.0412	.0803	.6003	1.4625	.6926	.2564
31.5119	.0435	.0714	.6232	1.4095	.7037	.2374
34.3295	.0461	.0648	.6431	1.3589	.7143	.2218
37.2606	.0494	.0592	.6618	1.3081	.7250	.2076
39.7971	.0525	.0553	.6763	1.2671	.7336	.1967
42.2430	.0559	.0521	.6887	1.2314	.7411	.1873
44.8722	.0597	.0493	.7000	1.1983	.7481	.1780
47.2518	.0633	.0471	.7085	1.1738	.7533	.1705
49.6688	.0670	.0452	.7154	1.1546	.7573	.1634
52.4321	.0710	.0434	.7212	1.1398	.7604	.1560
55.1003	.0748	.0419	.7249	1.1327	.7619	.1494
57.9239	.0783	.0406	.7269	1.1318	.7621	.1431
61.2556	.0821	.0392	.7274	1.1378	.7608	.1363
64.5759	.0852	.0380	.7264	1.1492	.7584	.1301
68.2753	.0882	.0369	.7242	1.1659	.7549	.1238
72.7361	.0911	.0358	.7206	1.1889	.7501	.1170
77.1408	.0935	.0348	.7168	1.2128	.7451	.1110
82.0203	.0955	.0339	.7123	1.2390	.7396	.1050
88.1527	.0975	.0329	.7070	1.2703	.7330	.0984
94.6840	.0991	.0320	.7019	1.3005	.7266	.0921
102.4017	.1005	.0312	.6967	1.3316	.7201	.0857
111.9224	.1018	.0304	.6914	1.3636	.7134	.0790
121.6304	.1028	.0298	.6871	1.3904	.7077	.0731
132.7553	.1036	.0292	.6832	1.4155	.7024	.0673
146.3975	.1042	.0287	.6795	1.4405	.6972	.0614
159.9183	.1046	.0284	.6765	1.4605	.6930	.0565
174.6035	.1048	.0281	.6741	1.4779	.6893	.0519
200.7772	.1049	.0278	.6711	1.5007	.6845	.0454

MACH NO = 3.50 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0429	.9871	1.1388	-.0614	1.0151	1.0075
1.0924	.0470	.9454	.9289	.0381	.9906	.9815
1.2904	.0505	.9098	.8127	.1158	.9716	.9586
1.7040	.0566	.8428	.6799	.2493	.9388	.9141
2.1205	.0614	.7837	.6127	.3586	.9119	.8733
2.7615	.0664	.7063	.5589	.4977	.8778	.8171
3.3785	.0694	.6441	.5328	.6073	.8509	.7695
4.2888	.0727	.5693	.5175	.7331	.8200	.7086
5.1339	.0749	.5137	.5146	.8225	.7980	.6600
6.3436	.0772	.4514	.5178	.9193	.7743	.6011
7.4391	.0788	.4077	.5247	.9832	.7585	.5561
8.6529	.0803	.3696	.5339	1.0357	.7457	.5136
10.3437	.0821	.3292	.5468	1.0878	.7329	.4641
11.8412	.0834	.3022	.5573	1.1208	.7248	.4276
13.4735	.0847	.2794	.5674	1.1477	.7181	.3938
15.7122	.0861	.2559	.5792	1.1747	.7115	.3554
17.6697	.0872	.2405	.5879	1.1921	.7073	.3274
19.7827	.0882	.2276	.5958	1.2065	.7037	.3018
22.0586	.0891	.2169	.6030	1.2185	.7008	.2783
24.5054	.0900	.2079	.6096	1.2285	.6983	.2568
27.8174	.0909	.1987	.6169	1.2390	.6957	.2325
30.6812	.0917	.1928	.6222	1.2460	.6940	.2150
33.7461	.0923	.1877	.6269	1.2520	.6925	.1989
37.0232	.0930	.1835	.6312	1.2573	.6912	.1841
40.5239	.0936	.1800	.6350	1.2620	.6901	.1706
45.2338	.0942	.1764	.6393	1.2671	.6888	.1553
49.2856	.0947	.1740	.6423	1.2708	.6879	.1442
53.6061	.0952	.1720	.6451	1.2741	.6871	.1339
58.2121	.0956	.1703	.6475	1.2772	.6864	.1245
64.3989	.0961	.1685	.6503	1.2806	.6855	.1137
69.7169	.0964	.1673	.6523	1.2832	.6849	.1059
75.3871	.0967	.1663	.6541	1.2857	.6843	.0986
81.4346	.0970	.1654	.6557	1.2880	.6837	.0919
87.8873	.0972	.1647	.6572	1.2902	.6832	.0856
96.5690	.0975	.1640	.6588	1.2928	.6825	.0785
104.0474	.0977	.1634	.6600	1.2948	.6820	.0732
112.0382	.0979	.1630	.6611	1.2966	.6816	.0683
120.5799	.0981	.1626	.6621	1.2984	.6811	.0637
129.7135	.0982	.1623	.6630	1.3001	.6807	.0595
142.0298	.0983	.1620	.6640	1.3020	.6803	.0546
152.6594	.0984	.1617	.6648	1.3034	.6799	.0509
164.0334	.0985	.1615	.6655	1.3047	.6796	.0476
176.2055	.0986	.1614	.6662	1.3059	.6793	.0444
192.6304	.0987	.1612	.6670	1.3072	.6790	.0408
203.1763	.0987	.1611	.6674	1.3080	.6788	.0387

MACH NO = 5.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0421	.9403	1.1388	-.0614	1.0151	1.0075
1.0437	.0447	.9084	.9659	.0176	.9957	.9873
1.2522	.0475	.8707	.8278	.1038	.9745	.9630
1.6071	.0516	.8123	.6939	.2273	.9442	.9242
2.0628	.0553	.7462	.6052	.3578	.9121	.8787
2.6198	.0581	.6765	.5481	.4906	.8795	.8289
3.2850	.0599	.6065	.5132	.6207	.8476	.7764
4.0646	.0611	.5387	.4949	.7418	.8178	.7226
4.9634	.0620	.4754	.4888	.8490	.7915	.6693
5.9847	.0628	.4180	.4922	.9382	.7696	.6175
7.1313	.0638	.3672	.5024	1.0078	.7525	.5681
8.4073	.0650	.3231	.5170	1.0581	.7399	.5216
9.8187	.0665	.2853	.5339	1.0948	.7312	.4784
11.7846	.0688	.2461	.5558	1.1225	.7244	.4289
13.9508	.0715	.2151	.5764	1.1375	.7207	.3850
16.3049	.0743	.1909	.5943	1.1457	.7187	.3464
18.3228	.0756	.1754	.6064	1.1504	.7175	.3190
21.0138	.0794	.1599	.6188	1.1560	.7161	.2886
23.8938	.0820	.1478	.6284	1.1626	.7145	.2619
26.9654	.0843	.1382	.6358	1.1702	.7126	.2383
29.5631	.0860	.1321	.6405	1.1770	.7110	.2215
32.9899	.0879	.1258	.6451	1.1860	.7088	.2026
36.6218	.0896	.1209	.6487	1.1952	.7065	.1858
39.6801	.0907	.1176	.6511	1.2025	.7047	.1737
43.7023	.0921	.1142	.6534	1.2115	.7025	.1600
47.9588	.0932	.1115	.6554	1.2200	.7004	.1476
52.4686	.0942	.1092	.6570	1.2281	.6984	.1365
56.2753	.0949	.1077	.6581	1.2343	.6969	.1283
61.3073	.0957	.1061	.6593	1.2416	.6951	.1189
66.6748	.0963	.1048	.6603	1.2485	.6934	.1102
72.4131	.0969	.1037	.6611	1.2550	.6918	.1023
77.2949	.0973	.1029	.6618	1.2600	.6906	.0964
83.7914	.0978	.1021	.6624	1.2658	.6892	.0895
90.7601	.0982	.1014	.6630	1.2713	.6878	.0831
96.7003	.0984	.1010	.6635	1.2754	.6868	.0784
104.6152	.0987	.1005	.6640	1.2801	.6856	.0728
113.1137	.0989	.1000	.6645	1.2844	.6846	.0677
122.2408	.0991	.0997	.6650	1.2883	.6836	.0629
130.0275	.0993	.0994	.6654	1.2912	.6829	.0593
140.4104	.0994	.0992	.6658	1.2944	.6821	.0552
151.5665	.0995	.0989	.6663	1.2974	.6814	.0513
163.5545	.0996	.0988	.6667	1.2999	.6808	.0477
173.7858	.0997	.0986	.6671	1.3018	.6803	.0450
187.4325	.0997	.0985	.6675	1.3039	.6798	.0418
202.0989	.0998	.0984	.6679	1.3057	.6794	.0389

MACH NO = 10.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0414	.9039	1.1388	-.0614	1.0151	1.0075
1.1055	.0442	.8612	.9166	.0452	.9889	.9799
1.3364	.0466	.8210	.7833	.1380	.9661	.9535
1.8118	.0498	.7467	.6362	.2977	.9269	.9032
2.2813	.0515	.6831	.5621	.4288	.8947	.8585
2.8387	.0522	.6177	.5119	.5618	.8620	.8108
3.6646	.0520	.5374	.4728	.7238	.8223	.7492
4.4288	.0513	.4768	.4553	.8443	.7927	.7000
5.2817	.0503	.4211	.4479	.9510	.7665	.6522
6.4639	.0492	.3596	.4498	1.0595	.7398	.5958
7.4938	.0485	.3171	.4580	1.1252	.7237	.5541
8.5901	.0480	.2804	.4703	1.1730	.7119	.5156
10.0428	.0478	.2418	.4894	1.2106	.7027	.4722
11.8900	.0481	.2046	.5150	1.2297	.6980	.4265
14.1683	.0494	.1711	.5455	1.2267	.6988	.3810
16.5636	.0514	.1457	.5746	1.2073	.7035	.3426
18.7033	.0536	.1287	.5972	1.1841	.7092	.3143
21.2975	.0568	.1132	.6204	1.1550	.7164	.2857
23.6042	.0599	.1026	.6374	1.1314	.7222	.2643
26.3912	.0638	.0929	.6537	1.1077	.7280	.2424
28.8617	.0673	.0861	.6650	1.0917	.7319	.2258
31.8412	.0715	.0798	.6752	1.0786	.7351	.2086
34.9366	.0755	.0748	.6827	1.0711	.7370	.1933
37.7050	.0789	.0713	.6874	1.0686	.7376	.1813
41.1187	.0827	.0678	.6912	1.0699	.7373	.1685
44.2586	.0858	.0654	.6932	1.0744	.7362	.1583
48.2503	.0892	.0629	.6942	1.0834	.7339	.1469
52.0215	.0919	.0611	.6940	1.0946	.7312	.1375
56.8809	.0946	.0592	.6925	1.1113	.7271	.1271
62.1013	.0968	.0577	.6902	1.1303	.7224	.1175
66.8760	.0983	.0567	.6879	1.1473	.7183	.1099
72.8383	.0997	.0556	.6850	1.1673	.7134	.1017
78.3171	.1006	.0549	.6825	1.1840	.7092	.0952
85.1757	.1013	.0541	.6797	1.2027	.7047	.0882
92.5744	.1019	.0535	.6771	1.2200	.7004	.0816
99.3764	.1022	.0530	.6751	1.2335	.6971	.0764
107.8878	.1025	.0526	.6732	1.2476	.6936	.0708
115.7114	.1026	.0523	.6718	1.2583	.6910	.0663
125.5002	.1027	.0519	.6704	1.2693	.6883	.0614
134.4969	.1027	.0517	.6695	1.2774	.6863	.0575
145.7528	.1027	.0515	.6687	1.2857	.6843	.0532
157.8880	.1027	.0513	.6681	1.2927	.6826	.0493
169.0421	.1026	.0512	.6677	1.2978	.6813	.0462
182.9998	.1025	.0510	.6675	1.3028	.6801	.0428
200.2966	.1024	.0509	.6674	1.3074	.6789	.0392

MACH NO = 15.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0412	.8970	1.1398	-.0614	1.0151	1.0075
1.1003	.0439	.8553	.9201	.0431	.9894	.9806
1.4095	.0467	.8021	.7513	.1657	.9593	.9454
1.7897	.0490	.7433	.6387	.2927	.9281	.9054
2.3744	.0507	.6649	.5478	.4564	.8879	.8502
2.9374	.0509	.6008	.5003	.5893	.8553	.8030
3.7637	.0502	.5225	.4624	.7511	.8155	.7425
4.5206	.0491	.4639	.4447	.8718	.7859	.6945
5.5785	.0475	.3978	.4353	1.0034	.7536	.6371
6.5048	.0462	.3513	.4359	1.0898	.7324	.5940
7.7482	.0449	.3014	.4441	1.1728	.7120	.5446
8.8000	.0440	.2677	.4549	1.2204	.7003	.5088
10.1722	.0433	.2321	.4717	1.2592	.6908	.4687
12.4669	.0430	.1881	.5023	1.2843	.6846	.4140
15.1432	.0437	.1525	.5376	1.2759	.6867	.3644
17.5660	.0451	.1296	.5669	1.2505	.6929	.3288
20.0002	.0471	.1124	.5933	1.2175	.7010	.2994
22.7258	.0499	.0980	.6191	1.1778	.7108	.2721
25.1219	.0529	.0884	.6386	1.1437	.7191	.2519
27.4866	.0551	.0808	.6552	1.1125	.7268	.2348
30.1127	.0601	.0742	.6707	1.0821	.7343	.2182
32.4281	.0637	.0696	.6819	1.0597	.7398	.2055
34.7460	.0675	.0658	.6911	1.0417	.7442	.1941
37.3925	.0717	.0623	.6993	1.0266	.7479	.1826
39.8210	.0755	.0597	.7049	1.0176	.7501	.1732
42.3672	.0793	.0574	.7089	1.0129	.7513	.1643
45.4349	.0834	.0552	.7118	1.0131	.7512	.1547
48.4054	.0869	.0534	.7127	1.0182	.7500	.1465
51.6528	.0902	.0519	.7123	1.0282	.7475	.1384
55.7086	.0935	.0503	.7103	1.0446	.7435	.1295
59.7737	.0961	.0490	.7074	1.0643	.7389	.1216
64.3861	.0984	.0478	.7036	1.0855	.7334	.1138
70.3417	.1004	.0466	.6986	1.1135	.7266	.1050
76.4406	.1018	.0456	.6937	1.1399	.7201	.0974
83.2033	.1029	.0448	.6889	1.1658	.7137	.0901
91.4507	.1037	.0440	.6842	1.1922	.7072	.0826
99.3886	.1041	.0435	.6805	1.2132	.7021	.0764
107.9402	.1044	.0430	.6774	1.2316	.6976	.0707
118.3527	.1045	.0426	.6745	1.2495	.6932	.0649
128.3712	.1045	.0423	.6724	1.2631	.6898	.0601
139.1694	.1045	.0421	.6708	1.2747	.6870	.0556
152.3292	.1044	.0419	.6694	1.2856	.6843	.0511
165.0026	.1042	.0417	.6685	1.2936	.6823	.0473
178.6723	.1041	.0416	.6679	1.3001	.6807	.0438
201.2228	.1038	.0414	.6674	1.3076	.6789	.0391

MACH NO = 20.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0412	.8943	1.1388	-.0614	1.0151	1.0075
1.0983	.0438	.8531	.9215	.0423	.9896	.9808
1.4049	.0465	.8003	.7528	.1642	.9597	.9459
1.7817	.0487	.7420	.6397	.2909	.9286	.9062
2.3601	.0503	.6643	.5482	.4540	.8885	.8514
2.9161	.0505	.6007	.4999	.5867	.8559	.8046
3.7306	.0497	.5230	.4608	.7490	.8161	.7447
4.4756	.0485	.4648	.4420	.8706	.7862	.6972
5.5145	.0467	.3990	.4310	1.0046	.7533	.6403
6.4220	.0453	.3528	.4302	1.0935	.7315	.5976
7.6365	.0437	.3032	.4366	1.1804	.7101	.5487
8.6605	.0427	.2695	.4460	1.2313	.6976	.5133
9.9914	.0418	.2340	.4613	1.2746	.6870	.4736
12.7676	.0411	.1809	.4972	1.3088	.6786	.4078
15.3314	.0414	.1479	.5305	1.3007	.6806	.3614
18.1832	.0427	.1221	.5646	1.2698	.6882	.3208
20.9831	.0447	.1039	.5944	1.2295	.6981	.2889
23.6931	.0474	.0909	.6199	1.1869	.7085	.2636
26.0408	.0501	.0822	.6396	1.1496	.7177	.2450
28.5487	.0535	.0748	.6582	1.1114	.7271	.2278
30.9652	.0572	.0692	.6738	1.0777	.7353	.2134
33.0828	.0607	.0652	.6855	1.0517	.7417	.2022
35.4033	.0648	.0615	.6963	1.0276	.7476	.1911
37.7292	.0689	.0586	.7050	1.0088	.7523	.1812
39.8657	.0726	.0563	.7111	.9962	.7554	.1730
42.3361	.0768	.0541	.7162	.9874	.7575	.1644
44.9662	.0809	.0522	.7195	.9843	.7583	.1561
47.5126	.0845	.0506	.7210	.9866	.7577	.1488
50.5683	.0882	.0491	.7209	.9946	.7557	.1410
53.9474	.0916	.0477	.7193	1.0082	.7524	.1332
57.7559	.0946	.0463	.7164	1.0269	.7478	.1254
61.5491	.0970	.0452	.7127	1.0472	.7426	.1185
66.2429	.0992	.0441	.7079	1.0726	.7366	.1109
71.6206	.1011	.0430	.7026	1.1004	.7298	.1033
77.2414	.1024	.0421	.6975	1.1267	.7233	.0964
84.5693	.1036	.0412	.6918	1.1565	.7160	.0887
92.9781	.1045	.0405	.6865	1.1847	.7091	.0813
101.4678	.1050	.0399	.6823	1.2078	.7034	.0749
111.7057	.1053	.0394	.6784	1.2301	.6979	.0685
122.8715	.1054	.0390	.6753	1.2492	.6932	.0626
135.0588	.1054	.0387	.6727	1.2653	.6893	.0572
146.9840	.1053	.0385	.6709	1.2774	.6863	.0528
161.3963	.1051	.0383	.6694	1.2886	.6836	.0483
177.1445	.1049	.0381	.6684	1.2975	.6814	.0442
201.6798	.1046	.0380	.6677	1.3067	.6791	.0390

MACH NO = 25.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = -3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0412	.8933	1.1388	-.0614	1.0151	1.0075
1.0975	.0437	.8522	.9221	.0419	.9897	.9809
1.4028	.0464	.7997	.7535	.1636	.9598	.9461
1.7780	.0486	.7416	.6402	.2900	.9288	.9066
2.3538	.0501	.6641	.5482	.4530	.8888	.8520
2.9069	.0503	.6007	.4996	.5857	.8562	.8054
3.7166	.0494	.5233	.4601	.7482	.8163	.7457
4.4564	.0482	.4652	.4407	.8703	.7863	.6984
5.4871	.0463	.3996	.4289	1.0053	.7531	.6417
6.3863	.0448	.3535	.4275	1.0955	.7310	.5992
7.5878	.0432	.3040	.4331	1.1841	.7092	.5505
8.5993	.0422	.2703	.4418	1.2367	.6963	.5153
9.9115	.0411	.2349	.4563	1.2820	.6852	.4758
12.9167	.0401	.1776	.4945	1.3214	.6755	.4047
15.7027	.0404	.1429	.5302	1.3115	.6779	.3555
18.7344	.0416	.1168	.5658	1.2771	.6864	.3140
21.6620	.0437	.0990	.5963	1.2335	.6971	.2821
24.4443	.0463	.0865	.6223	1.1880	.7083	.2573
26.8379	.0491	.0782	.6424	1.1479	.7181	.2392
29.3318	.0526	.0714	.6613	1.1072	.7281	.2229
31.7052	.0563	.0662	.6771	1.0713	.7369	.2093
33.7872	.0599	.0625	.6892	1.0431	.7439	.1987
36.0315	.0639	.0591	.7002	1.0172	.7502	.1884
38.2710	.0681	.0564	.7091	.9968	.7552	.1791
40.3441	.0719	.0543	.7155	.9830	.7586	.1713
42.7164	.0761	.0523	.7207	.9733	.7610	.1631
45.2435	.0803	.0505	.7240	.9696	.7619	.1553
47.6913	.0839	.0490	.7254	.9716	.7614	.1484
50.5989	.0876	.0476	.7252	.9795	.7595	.1409
53.8206	.0911	.0462	.7235	.9931	.7561	.1335
57.4022	.0941	.0449	.7204	1.0117	.7516	.1261
60.9353	.0965	.0439	.7167	1.0317	.7466	.1195
65.2468	.0987	.0428	.7119	1.0566	.7405	.1124
70.1752	.1006	.0417	.7065	1.0838	.7339	.1053
75.3852	.1021	.0408	.7013	1.1101	.7274	.0986
81.9709	.1034	.0399	.6957	1.1392	.7202	.0913
89.3306	.1044	.0391	.6905	1.1665	.7136	.0844
96.9476	.1051	.0385	.6861	1.1898	.7078	.0782
106.7590	.1056	.0379	.6817	1.2143	.7018	.0715
118.1236	.1059	.0375	.6777	1.2369	.6963	.0650
130.5996	.1059	.0371	.6745	1.2561	.6916	.0591
143.0000	.1058	.0369	.6722	1.2707	.6880	.0542
157.9190	.1057	.0367	.6703	1.2840	.6847	.0493
174.3081	.1054	.0365	.6690	1.2945	.6821	.0449
201.0785	.1051	.0363	.6679	1.3056	.6794	.0391

MACH NO = 30.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 3.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8781	.0411	.8927	1.1388	-.0614	1.0151	1.0075
1.0970	.0437	.8517	.9224	.0417	.9898	.9810
1.4017	.0464	.7993	.7539	.1632	.9599	.9462
1.7760	.0485	.7413	.6404	.2895	.9289	.9068
2.3503	.0500	.6639	.5483	.4524	.8889	.8523
3.0528	.0501	.5849	.4899	.6181	.8482	.7939
3.7088	.0493	.5234	.4596	.7477	.8164	.7462
4.6421	.0476	.4516	.4366	.8990	.7792	.6874
5.6898	.0457	.3879	.4268	1.0299	.7471	.6316
6.5991	.0443	.3433	.4265	1.1165	.7258	.5900
7.8083	.0426	.2955	.4329	1.2009	.7051	.5425
8.8229	.0416	.2632	.4421	1.2506	.6929	.5081
10.1326	.0406	.2291	.4567	1.2930	.6825	.4697
13.3926	.0395	.1701	.4980	1.3291	.6736	.3954
16.4138	.0399	.1352	.5360	1.3131	.6775	.3448
19.3684	.0412	.1118	.5598	1.2769	.6864	.3065
22.4410	.0434	.0945	.6011	1.2296	.6980	.2747
25.0862	.0459	.0834	.6254	1.1850	.7090	.2522
27.5653	.0489	.0753	.6462	1.1421	.7195	.2342
30.1012	.0525	.0688	.6654	1.0992	.7301	.2183
32.3019	.0561	.0642	.6803	1.0645	.7386	.2061
34.6086	.0602	.0603	.6938	1.0322	.7465	.1948
36.6767	.0640	.0575	.7039	1.0077	.7525	.1856
38.7414	.0680	.0551	.7122	.9893	.7573	.1773
41.0408	.0723	.0529	.7192	.9728	.7611	.1688
43.2439	.0763	.0511	.7238	.9642	.7632	.1614
45.7996	.0806	.0493	.7269	.9610	.7640	.1537
48.2928	.0843	.0479	.7280	.9640	.7633	.1467
50.9933	.0878	.0466	.7275	.9723	.7612	.1399
54.2648	.0913	.0452	.7254	.9871	.7576	.1325
57.5153	.0940	.0441	.7223	1.0049	.7532	.1258
61.3650	.0966	.0429	.7180	1.0276	.7476	.1188
65.2941	.0987	.0419	.7134	1.0510	.7419	.1123
69.7581	.1004	.0409	.7084	1.0764	.7357	.1058
75.4581	.1021	.0399	.7025	1.1060	.7284	.0985
81.3088	.1033	.0391	.6973	1.1324	.7219	.0920
88.3523	.1044	.0383	.6921	1.1592	.7153	.0852
95.6694	.1051	.0377	.6878	1.1824	.7096	.0792
104.1653	.1057	.0372	.6837	1.2048	.7041	.0731
115.3461	.1061	.0367	.6794	1.2287	.6983	.0665
127.2603	.1062	.0363	.6760	1.2488	.6933	.0606
141.5650	.1061	.0360	.6730	1.2572	.6888	.0547
155.9973	.1060	.0358	.6709	1.2810	.6854	.0499
171.8190	.1057	.0356	.6694	1.2921	.6827	.0455
201.0670	.1054	.0354	.6681	1.3048	.6796	.0391

MACH NO = 3.50 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0425	.9908	1.1617	-.0703	1.0198	1.0098
1.0699	.0465	.9447	.9486	.0270	.9924	.9807
1.2624	.0499	.9058	.8310	.1019	.9714	.9554
1.5741	.0546	.8488	.7186	.2031	.9429	.9170
2.0641	.0603	.7714	.6294	.3299	.9073	.8625
2.5450	.0641	.7072	.5841	.4314	.8788	.8150
3.1111	.0671	.6434	.5548	.5300	.8510	.7654
3.7668	.0697	.5822	.5392	.6205	.8256	.7150
4.7181	.0725	.5117	.5330	.7190	.7979	.6526
5.5880	.0745	.4614	.5345	.7860	.7791	.6044
6.5579	.0762	.4170	.5401	.8420	.7633	.5584
7.6306	.0780	.3784	.5484	.8873	.7506	.5150
9.1204	.0801	.3378	.5606	.9316	.7381	.4649
10.4354	.0816	.3108	.5705	.9594	.7303	.4281
12.2387	.0835	.2831	.5824	.9869	.7226	.3862
13.8135	.0849	.2649	.5912	1.0046	.7176	.3558
15.9542	.0865	.2464	.6011	1.0227	.7125	.3214
17.8101	.0876	.2344	.6082	1.0347	.7092	.2965
20.3172	.0889	.2222	.6160	1.0474	.7056	.2685
23.0426	.0901	.2125	.6228	1.0580	.7026	.2434
25.3883	.0909	.2062	.6276	1.0653	.7006	.2254
28.5372	.0919	.1997	.6329	1.0733	.6983	.2049
31.2394	.0926	.1955	.6367	1.0789	.6967	.1901
34.8594	.0933	.1912	.6409	1.0851	.6950	.1734
37.9603	.0939	.1883	.6439	1.0895	.6938	.1612
42.1096	.0945	.1854	.6472	1.0944	.6924	.1473
45.6617	.0950	.1834	.6496	1.0980	.6914	.1372
50.4148	.0955	.1814	.6522	1.1020	.6902	.1257
54.4862	.0958	.1801	.6542	1.1050	.6894	.1173
59.9397	.0962	.1787	.6563	1.1084	.6885	.1076
65.8333	.0966	.1775	.6582	1.1115	.6876	.0988
70.8931	.0968	.1767	.6596	1.1139	.6869	.0923
77.6871	.0971	.1759	.6611	1.1166	.6861	.0848
83.5283	.0973	.1754	.6623	1.1187	.6856	.0793
91.3812	.0975	.1748	.6636	1.1211	.6849	.0729
98.1394	.0976	.1744	.6645	1.1228	.6844	.0682
107.2318	.0978	.1740	.6656	1.1248	.6838	.0627
115.0609	.0979	.1738	.6664	1.1262	.6834	.0587
125.5979	.0980	.1735	.6674	1.1279	.6830	.0540
134.6734	.0980	.1733	.6681	1.1290	.6827	.0505
146.9905	.0981	.1731	.6689	1.1303	.6823	.0465
160.1557	.0982	.1729	.6697	1.1315	.6820	.0428
171.5833	.0982	.1728	.6703	1.1323	.6817	.0400
186.9692	.0983	.1727	.6709	1.1332	.6815	.0368
200.2248	.0983	.1726	.6715	1.1338	.6813	.0345

MACH NO = 5.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0418	.9440	1.1617	-.0703	1.0198	1.0098
1.0212	.0443	.9088	.9872	.0065	.9982	.9874
1.2229	.0471	.8678	.8476	.0895	.9748	.9605
1.5646	.0510	.8049	.7125	.2065	.9420	.9181
1.9934	.0543	.7361	.6243	.3257	.9084	.8700
2.5044	.0569	.6661	.5685	.4424	.8756	.8188
3.1082	.0585	.5967	.5338	.5547	.8441	.7656
3.8090	.0597	.5305	.5154	.6571	.8153	.7119
4.6093	.0606	.4693	.5088	.7461	.7903	.6592
5.5099	.0615	.4143	.5114	.8190	.7698	.6084
6.5113	.0626	.3660	.5205	.8749	.7541	.5604
7.6144	.0639	.3243	.5336	.9156	.7427	.5156
8.8210	.0655	.2887	.5489	.9435	.7348	.4742
10.8337	.0683	.2454	.5726	.9678	.7280	.4181
12.7104	.0711	.2170	.5913	.9781	.7251	.3766
14.7755	.0741	.1944	.6077	.9839	.7234	.3395
16.9912	.0771	.1769	.6211	.9883	.7222	.3070
19.3507	.0799	.1633	.6316	.9931	.7209	.2786
21.8537	.0825	.1526	.6398	.9989	.7192	.2538
25.0484	.0853	.1429	.6469	1.0073	.7169	.2278
27.8727	.0873	.1366	.6513	1.0152	.7147	.2089
30.8483	.0891	.1316	.6547	1.0234	.7124	.1921
33.9819	.0906	.1276	.6572	1.0317	.7100	.1771
37.2834	.0920	.1243	.6592	1.0398	.7077	.1637
40.7674	.0932	.1217	.6608	1.0477	.7055	.1515
44.4537	.0942	.1195	.6621	1.0553	.7034	.1405
49.1778	.0952	.1174	.6633	1.0639	.7009	.1285
53.3955	.0960	.1160	.6642	1.0708	.6990	.1194
57.8964	.0966	.1148	.6648	1.0774	.6972	.1110
62.7066	.0972	.1138	.6654	1.0836	.6954	.1033
67.8519	.0976	.1130	.6658	1.0895	.6937	.0961
73.3592	.0980	.1122	.6662	1.0951	.6922	.0895
80.4856	.0984	.1115	.6667	1.1012	.6905	.0821
86.8915	.0986	.1110	.6670	1.1059	.6892	.0764
93.7567	.0988	.1106	.6674	1.1101	.6880	.0712
101.1163	.0990	.1103	.6677	1.1139	.6869	.0663
109.0071	.0991	.1100	.6681	1.1174	.6859	.0618
117.4689	.0992	.1097	.6685	1.1205	.6851	.0575
126.5438	.0993	.1095	.6689	1.1232	.6843	.0536
138.3068	.0993	.1093	.6693	1.1261	.6835	.0492
148.8939	.0994	.1091	.6697	1.1282	.6829	.0459
160.2501	.0994	.1090	.6701	1.1300	.6824	.0428
172.4318	.0994	.1089	.6705	1.1316	.6819	.0398
185.4993	.0994	.1088	.6709	1.1329	.6816	.0371
202.4412	.0994	.1087	.6714	1.1343	.6812	.0341

MACH NO = 10.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0411	.9076	1.1617	-.0703	1.0198	1.0098
1.0777	.0439	.8612	.9399	.0317	.9911	.9797
1.2934	.0460	.8188	.8080	.1182	.9668	.9514
1.7325	.0490	.7419	.6610	.2639	.9258	.8987
2.1608	.0506	.6772	.5867	.3807	.8930	.8525
2.6639	.0513	.6118	.5361	.4968	.8604	.8041
3.4017	.0512	.5324	.4966	.6355	.8214	.7422
4.0779	.0506	.4732	.4789	.7366	.7929	.6933
4.8264	.0498	.4191	.4710	.8249	.7681	.6462
5.8541	.0489	.3598	.4721	.9133	.7433	.5910
6.7413	.0484	.3190	.4793	.9661	.7285	.5504
7.6781	.0482	.2839	.4905	1.0039	.7178	.5132
8.9078	.0482	.2470	.5079	1.0332	.7096	.4714
10.7182	.0490	.2062	.5353	1.0483	.7053	.4209
12.8859	.0508	.1717	.5668	1.0413	.7073	.3731
14.8510	.0531	.1491	.5924	1.0237	.7123	.3382
16.8695	.0559	.1318	.6152	1.0018	.7184	.3086
19.2333	.0597	.1167	.6373	.9765	.7255	.2799
21.3471	.0633	.1066	.6531	.9568	.7311	.2584
23.5924	.0671	.0985	.6660	.9407	.7356	.2397
26.0213	.0715	.0913	.6775	.9271	.7394	.2209
28.2877	.0753	.0863	.6851	.9194	.7416	.2064
30.6366	.0791	.0822	.6907	.9155	.7427	.1932
33.4686	.0832	.0784	.6952	.9151	.7428	.1794
36.1275	.0866	.0756	.6977	.9184	.7419	.1682
39.4443	.0903	.0729	.6988	.9263	.7396	.1559
42.6568	.0932	.0709	.6984	.9370	.7366	.1457
46.2296	.0957	.0692	.6969	.9511	.7327	.1358
50.8534	.0982	.0674	.6940	.9705	.7272	.1247
55.3716	.0998	.0662	.6908	.9894	.7219	.1156
60.2312	.1010	.0651	.6875	1.0084	.7166	.1071
66.2449	.1020	.0641	.6837	1.0293	.7107	.0982
71.9436	.1025	.0634	.6806	1.0464	.7059	.0911
78.0816	.1029	.0628	.6779	1.0619	.7015	.0844
85.6760	.1031	.0623	.6752	1.0776	.6971	.0775
92.8682	.1031	.0619	.6734	1.0895	.6938	.0718
101.7647	.1031	.0615	.6717	1.1011	.6905	.0659
110.1886	.1030	.0613	.6706	1.1096	.6881	.0611
119.2575	.1029	.0611	.6698	1.1168	.6861	.0567
130.4761	.1027	.0608	.6692	1.1235	.6842	.0521
141.1014	.1026	.0607	.6690	1.1282	.6829	.0483
152.5438	.1024	.0606	.6689	1.1320	.6818	.0448
166.7035	.1023	.0605	.6690	1.1353	.6809	.0412
180.1186	.1021	.0604	.6693	1.1375	.6803	.0382
201.0989	.1019	.0603	.6698	1.1397	.6797	.0343

MACH NO = 15.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0410	.9006	1.1617	-.0703	1.0198	1.0098
1.0727	.0436	.8554	.9435	.0297	.9917	.9803
1.2834	.0456	.8139	.8121	.1149	.9677	.9527
1.7118	.0483	.7386	.6638	.2593	.9271	.9010
2.1291	.0496	.6751	.5876	.3757	.8944	.8558
2.7517	.0501	.5949	.5248	.5205	.8537	.7962
3.3322	.0497	.5327	.4928	.6317	.8224	.7476
4.1574	.0486	.4604	.4690	.7594	.7866	.6880
4.8909	.0475	.4084	.4603	.8478	.7617	.6424
5.8875	.0462	.3515	.4596	.9376	.7365	.5894
6.7383	.0453	.3125	.4650	.9923	.7211	.5506
7.6274	.0447	.2788	.4743	1.0328	.7097	.5151
8.7807	.0442	.2434	.4894	1.0658	.7004	.4754
10.9362	.0443	.1949	.5212	1.0880	.6942	.4156
13.1455	.0454	.1607	.5538	1.0794	.6966	.3681
15.6119	.0475	.1339	.5871	1.0521	.7043	.3264
17.8084	.0502	.1168	.6130	1.0218	.7128	.2966
19.9614	.0533	.1041	.6351	.9912	.7214	.2721
22.2922	.0571	.0937	.6554	.9596	.7303	.2499
24.3383	.0609	.0866	.6705	.9347	.7373	.2331
26.3492	.0648	.0810	.6829	.9137	.7432	.2187
28.3468	.0689	.0766	.6930	.8966	.7480	.2061
30.5860	.0734	.0726	.7018	.8824	.7520	.1935
32.6568	.0775	.0697	.7078	.8739	.7544	.1832
34.8211	.0815	.0672	.7121	.8697	.7555	.1735
37.1243	.0855	.0650	.7148	.8699	.7555	.1643
39.9091	.0896	.0629	.7158	.8754	.7539	.1544
42.6612	.0929	.0612	.7151	.8853	.7512	.1457
45.7084	.0960	.0597	.7130	.8994	.7472	.1371
49.5467	.0988	.0582	.7092	.9196	.7415	.1277
53.4639	.1009	.0570	.7048	.9412	.7354	.1193
57.9051	.1026	.0558	.6998	.9649	.7288	.1110
63.0309	.1038	.0548	.6945	.9899	.7218	.1028
69.7625	.1047	.0539	.6884	1.0182	.7138	.0937
76.6323	.1051	.0531	.6835	1.0418	.7072	.0859
84.0964	.1053	.0526	.6794	1.0624	.7014	.0788
92.2040	.1053	.0521	.6761	1.0800	.6964	.0723
102.0372	.1051	.0517	.6732	1.0961	.6919	.0657
111.7007	.1049	.0515	.6713	1.1080	.6886	.0604
122.2092	.1047	.0513	.6700	1.1175	.6859	.0554
134.9716	.1044	.0511	.6691	1.1258	.6836	.0504
147.5280	.1041	.0510	.6686	1.1314	.6820	.0463
161.1936	.1039	.0508	.6685	1.1357	.6808	.0425
176.0683	.1036	.0508	.6687	1.1387	.6799	.0390
201.8677	.1033	.0506	.6693	1.1415	.6791	.0342

MACH NO = 20.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0409	.8979	1.1617	-.0703	1.0198	1.0098
1.0709	.0434	.8532	.9449	.0290	.9919	.9806
1.3568	.0460	.7977	.7778	.1422	.9600	.9434
1.7043	.0480	.7373	.6649	.2576	.9276	.9019
2.2317	.0494	.6583	.5729	.4029	.8868	.8454
2.7334	.0496	.5946	.5244	.5186	.8542	.7978
3.4612	.0489	.5179	.4853	.6570	.8153	.7376
4.1204	.0479	.4610	.4665	.7588	.7867	.6904
5.0315	.0464	.3971	.4555	.8690	.7557	.6343
5.8204	.0453	.3526	.4545	.9409	.7355	.5927
6.8671	.0441	.3048	.4605	1.0099	.7161	.5451
7.7424	.0433	.2725	.4693	1.0495	.7050	.5109
8.8712	.0427	.2385	.4837	1.0822	.6958	.4726
11.4325	.0426	.1835	.5210	1.1057	.6892	.4039
14.0196	.0438	.1474	.5586	1.0897	.6937	.3522
16.3320	.0457	.1249	.5890	1.0606	.7019	.3160
18.7861	.0485	.1076	.6174	1.0240	.7122	.2849
21.1229	.0518	.0954	.6411	.9878	.7224	.2606
23.3419	.0556	.0866	.6607	.9542	.7318	.2410
25.4615	.0597	.0800	.6770	.9245	.7402	.2248
27.5110	.0640	.0749	.6904	.8993	.7472	.2112
29.3435	.0680	.0713	.7003	.8806	.7525	.2003
31.3628	.0724	.0680	.7050	.8647	.7569	.1895
33.4281	.0769	.0653	.7156	.8539	.7600	.1796
35.5889	.0813	.0630	.7201	.8484	.7615	.1703
37.9013	.0856	.0610	.7226	.8483	.7615	.1614
40.1610	.0893	.0593	.7233	.8531	.7602	.1535
42.8704	.0930	.0577	.7223	.8634	.7573	.1450
45.8966	.0963	.0562	.7197	.8789	.7530	.1366
49.2369	.0990	.0549	.7157	.8985	.7474	.1284
52.9567	.1013	.0537	.7108	.9213	.7411	.1203
57.2045	.1030	.0525	.7052	.9465	.7340	.1122
61.6791	.1043	.0516	.6997	.9709	.7271	.1048
67.2347	.1053	.0506	.6938	.9975	.7196	.0969
73.5527	.1059	.0498	.6883	1.0228	.7125	.0892
80.9320	.1063	.0492	.6833	1.0468	.7058	.0817
89.7492	.1063	.0486	.6787	1.0694	.6994	.0742
98.6294	.1062	.0483	.6754	1.0868	.6945	.0679
108.8656	.1060	.0480	.6727	1.1020	.6903	.0618
119.7630	.1057	.0478	.6707	1.1139	.6869	.0565
131.5379	.1053	.0476	.6695	1.1231	.6843	.0517
144.4246	.1050	.0475	.6687	1.1302	.6823	.0472
158.6749	.1047	.0474	.6685	1.1353	.6809	.0432
173.0487	.1045	.0473	.6685	1.1385	.6800	.0397
201.3361	.1041	.0471	.6692	1.1418	.6791	.0343

MACH NO = 25.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8604	.0409	.8969	1.1617	-.0703	1.0198	1.0098
1.0700	.0434	.8523	.9455	.0286	.9920	.9807
1.3548	.0459	.7970	.7785	.1416	.9602	.9437
1.7008	.0479	.7369	.6654	.2568	.9278	.9023
2.2258	.0493	.6582	.5730	.4020	.8870	.8459
2.7249	.0494	.5947	.5242	.5177	.8545	.7986
3.4482	.0487	.5181	.4846	.6563	.8155	.7386
4.1930	.0476	.4613	.4654	.7585	.7868	.6916
5.0070	.0461	.3977	.4536	.8696	.7555	.6357
5.7888	.0449	.3532	.4521	.9424	.7351	.5942
6.8247	.0436	.3055	.4573	1.0128	.7153	.5469
7.6897	.0428	.2732	.4656	1.0537	.7038	.5128
8.8034	.0421	.2393	.4793	1.0880	.6942	.4747
11.5534	.0417	.1803	.5189	1.1147	.6867	.4011
14.3087	.0429	.1429	.5586	1.0963	.6918	.3472
16.9773	.0450	.1185	.5931	1.0611	.7018	.3072
19.7067	.0482	.1010	.6239	1.0185	.7137	.2748
22.0585	.0517	.0900	.6472	.9802	.7245	.2519
24.2624	.0556	.0820	.6667	.9449	.7344	.2337
26.3487	.0598	.0761	.6828	.9139	.7431	.2187
28.3557	.0642	.0716	.6960	.8878	.7504	.2060
30.3251	.0687	.0680	.7066	.8670	.7563	.1949
32.3007	.0733	.0651	.7149	.8514	.7607	.1849
34.3305	.0778	.0627	.7209	.8413	.7635	.1756
36.4688	.0823	.0606	.7249	.8358	.7648	.1668
38.9528	.0869	.0586	.7269	.8384	.7644	.1576
41.4358	.0919	.0570	.7268	.8458	.7623	.1494
44.1856	.0944	.0555	.7249	.8587	.7586	.1413
47.1876	.0975	.0541	.7215	.8761	.7537	.1333
50.4690	.1000	.0529	.7170	.8970	.7479	.1256
54.1535	.1021	.0517	.7117	.9206	.7412	.1179
58.4018	.1037	.0506	.7058	.9465	.7340	.1102
63.2138	.1050	.0496	.6998	.9728	.7266	.1025
69.0081	.1059	.0487	.6938	.9999	.7189	.0946
75.1276	.1065	.0480	.6886	1.0237	.7123	.0875
82.3086	.1068	.0474	.6837	1.0466	.7058	.0804
90.9475	.1069	.0469	.6792	1.0685	.6997	.0733
100.4693	.1068	.0465	.6756	1.0872	.6944	.0667
110.3875	.1065	.0463	.6729	1.1020	.6903	.0610
120.8887	.1061	.0461	.6709	1.1137	.6870	.0560
132.1623	.1058	.0459	.6696	1.1229	.6844	.0514
145.4636	.1054	.0458	.6688	1.1304	.6823	.0469
158.9491	.1051	.0457	.6685	1.1353	.6809	.0431
173.8004	.1049	.0456	.6685	1.1387	.6799	.0395
200.7115	.1046	.0455	.6692	1.1418	.6791	.0344

MACH NO = 30.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0469	.8962	1.1617	-.0703	1.0198	1.0098
1.0696	.0434	.8518	.9458	.0284	.9920	.9808
1.3537	.0458	.7966	.7789	.1413	.9603	.9438
1.6989	.0478	.7356	.6656	.2563	.9280	.9025
2.2226	.0492	.6580	.5731	.4015	.8872	.8463
2.7202	.0493	.5946	.5241	.5172	.8546	.7990
3.4412	.0486	.5182	.4842	.6559	.8156	.7391
4.0936	.0475	.4615	.4648	.7583	.7868	.6922
4.9937	.0459	.3979	.4526	.8699	.7555	.6365
5.7716	.0447	.3535	.4507	.9432	.7349	.5951
6.8016	.0433	.3059	.4555	1.0144	.7149	.5479
7.6609	.0425	.2736	.4636	1.0559	.7032	.5139
8.7664	.0417	.2397	.4769	1.0911	.6933	.4759
11.7193	.0413	.1770	.5192	1.1196	.6853	.3974
14.6581	.0425	.1385	.5611	1.0980	.6914	.3414
17.4708	.0448	.1142	.5968	1.0594	.7022	.3008
20.0970	.0479	.0982	.6260	1.0173	.7141	.2707
22.5253	.0515	.0873	.6502	.9765	.7255	.2478
24.7792	.0556	.0796	.6701	.9391	.7360	.2298
26.8994	.0600	.0739	.6866	.9066	.7452	.2151
28.9325	.0646	.0696	.6999	.8796	.7528	.2026
30.9267	.0693	.0662	.7105	.8583	.7587	.1917
32.9328	.0740	.0635	.7186	.8429	.7631	.1819
35.0063	.0788	.0611	.7243	.8334	.7657	.1727
37.2058	.0834	.0591	.7279	.8301	.7667	.1640
39.5522	.0877	.0573	.7292	.8330	.7659	.1556
42.1084	.0917	.0558	.7286	.8420	.7633	.1473
44.9323	.0952	.0543	.7260	.8567	.7592	.1392
48.2174	.0984	.0529	.7218	.8772	.7534	.1308
51.5949	.1008	.0516	.7168	.8995	.7472	.1231
55.4334	.1027	.0505	.7111	.9246	.7401	.1155
59.8615	.1043	.0494	.7050	.9513	.7326	.1077
64.7092	.1054	.0484	.6991	.9771	.7254	.1004
70.0920	.1062	.0476	.6936	1.0016	.7185	.0933
76.2534	.1068	.0469	.6886	1.0249	.7119	.0863
83.5447	.1072	.0464	.6838	1.0475	.7056	.0793
92.2333	.1072	.0459	.6793	1.0692	.6995	.0723
101.7877	.1071	.0456	.6757	1.0878	.6943	.0659
111.7704	.1068	.0453	.6729	1.1026	.6901	.0603
122.3242	.1064	.0451	.6709	1.1144	.6868	.0554
133.6319	.1060	.0450	.6696	1.1235	.6842	.0509
145.8643	.1057	.0449	.6688	1.1303	.6823	.0468
159.1896	.1054	.0448	.6685	1.1352	.6809	.0430
173.7900	.1052	.0447	.6686	1.1386	.6800	.0395
200.6168	.1048	.0446	.6692	1.1417	.6791	.0344

MACH NO = 3.50 CONE ANGLE = 9.00 ANGLE OF ATTACK = 2.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8476	.0422	.9950	1.1854	-.0792	1.0251	1.0125
1.0476	.0461	.9448	.9790	.0159	.9950	.9804
1.2348	.0494	.9170	.8489	.0883	.9720	.9527
1.5367	.0539	.8424	.7367	.1846	.9415	.9112
1.9155	.0583	.7741	.6618	.2796	.9114	.8651
2.3470	.0621	.7130	.6116	.3728	.8819	.8157
2.8677	.0651	.6569	.5774	.4628	.8534	.7644
3.4697	.0677	.5949	.5595	.5445	.8275	.7124
4.1564	.0700	.5275	.5519	.6158	.8049	.6612
4.9296	.0721	.4758	.5511	.6768	.7856	.6117
5.7912	.0740	.4333	.5549	.7276	.7695	.5645
6.7428	.0758	.3979	.5620	.7683	.7566	.5203
8.0613	.0781	.3636	.5731	.8075	.7442	.4693
9.2214	.0799	.3324	.5834	.8318	.7365	.4320
10.8051	.0820	.2945	.5976	.8558	.7249	.3898
12.5445	.0839	.2723	.6179	.8746	.7230	.3620
14.4415	.0857	.2547	.6429	.8899	.7191	.3383
16.5025	.0873	.2409	.6727	.9026	.7141	.2384
18.7347	.0887	.2296	.6974	.9134	.7107	.2617
21.1456	.0899	.2212	.7232	.9227	.7077	.2379
23.7421	.0910	.2142	.7392	.9309	.7051	.2187
26.5331	.0921	.2087	.7625	.9380	.7029	.1978
29.5247	.0928	.2043	.7862	.9444	.7018	.1868
32.7580	.0936	.2007	.8035	.9501	.6990	.1655
36.2235	.0941	.1984	.8218	.9542	.6977	.1544
39.9270	.0947	.1961	.8344	.9589	.6963	.1417
43.9231	.0952	.1940	.8468	.9631	.6949	.1301
47.3787	.0956	.1924	.8538	.9670	.6937	.1197
51.9490	.0961	.1911	.8617	.9705	.6926	.1101
56.9925	.0963	.1901	.8623	.9734	.6915	.1014
62.2329	.0966	.1891	.8638	.9768	.6906	.0934
68.0074	.0968	.1883	.8631	.9795	.6897	.0860
74.2554	.0971	.1877	.8664	.9821	.6889	.0793
81.1183	.0971	.1872	.8675	.9842	.6882	.0731
88.7407	.0973	.1868	.8685	.9862	.6876	.0674
96.6374	.0974	.1865	.8693	.9876	.6872	.0631
105.9861	.0975	.1862	.8713	.9892	.6866	.0582
116.2421	.0976	.1860	.8711	.9906	.6862	.0537
127.1617	.0976	.1858	.8719	.9919	.6858	.0495
139.0981	.0977	.1856	.8727	.9929	.6855	.0457
151.9525	.0977	.1855	.8734	.9939	.6852	.0421
165.7201	.0978	.1853	.8741	.9947	.6849	.0388
180.8431	.0978	.1852	.8747	.9954	.6847	.0358
195.6631	.0978	.1852	.8753	.9960	.6845	.0330
211.7175	.0978	.1851	.8759	.9965	.6843	.0305

MACH NO = 6.[illegible] CONE ANGLE = 9.0 ANGLE OF ATTACK = 5.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.9476	.2415	.9491	1.1854	-.0792	1.0251	1.0125
.9989	.2440	.9[illegible]	1.0[illegible]33	-.0046	1.0015	.9879
1.1940	.2467	.8679	.8681	.0755	.9761	.9586
1.5215	.2503	.7995	.7322	.1861	.9411	.9132
1.9195	.2535	.7298	.6472	.2979	.9079	.8536
2.3366	.2553	.6[illegible]	.5836	.3975	.8741	.8116
2.9342	.2574	.5917	.5548	.4952	.8431	.7583
3.5679	.2585	.5277	.5359	.5829	.8153	.7050
4.3769	.2605	.4682	.5236	.6532	.7915	.6550
5.0716	.2615	.4154	.5310	.7191	.7722	.6034
5.9459	.2615	.3503	.5376	.7654	.7575	.5569
6.9336	.2633	.3216	.5492	.7989	.7449	.5136
7.9344	.2645	.2957	.5628	.8218	.7337	.4778
9.6351	.2675	.2546	.5841	.8417	.7314	.4201
11.5232	.2703	.2229	.6042	.8512	.7304	.3732
13.2572	.2723	.2023	.6147	.8554	.7290	.3387
15.5251	.2774	.1829	.6321	.8597	.7277	.3018
17.5824	.2803	.1704	.6424	.8639	.7263	.2748
20.1956	.2875	.1590	.6556	.8725	.7243	.2467
22.9637	.2862	.1515	.6565	.8783	.7218	.2226
25.4156	.2832	.1450	.6639	.8856	.7195	.2050
28.4856	.2912	.1396	.6600	.8947	.7166	.1864
31.1946	.2917	.1365	.5648	.9023	.7140	.1726
34.6364	.2932	.1333	.5664	.9112	.7114	.1577
37.6850	.2942	.1312	.5674	.9195	.7091	.1466
41.5934	.2953	.1291	.5683	.9269	.7064	.1344
45.2350	.2952	.1274	.5689	.9350	.7038	.1233
49.5541	.2961	.1262	.5693	.9415	.7018	.1149
54.4000	.2974	.1251	.5696	.9489	.6994	.1056
58.7411	.2978	.1243	.5698	.9547	.6976	.0984
64.7457	.2982	.1235	.5700	.9611	.6955	.0915
69.3734	.2985	.1231	.5702	.9661	.6941	.0844
75.9756	.2987	.1224	.5704	.9714	.6923	.0777
82.0271	.2988	.1220	.5707	.9761	.6908	.0715
89.5537	.2989	.1217	.5710	.9796	.6897	.0667
97.4516	.2990	.1214	.5713	.9832	.6885	.0614
104.9177	.2991	.1211	.5716	.9859	.6877	.0573
114.3314	.2991	.1210	.5720	.9887	.6864	.0527
122.9854	.2991	.1208	.5724	.9907	.6862	.0492
133.0541	.2991	.1206	.5729	.9927	.6855	.0453
145.0716	.2992	.1205	.5734	.9944	.6851	.0417
156.7606	.2992	.1204	.5738	.9955	.6847	.0389
170.7470	.2992	.1203	.5743	.9966	.6843	.0358
183.731	.299[illegible]	.1203	.5748	.9974	.6841	.0335
202.3077	.2991	.1202	.5754	.9982	.6833	.0314

MACH NO = 10.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 3.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0409	.9116	1.1854	-.0792	1.0251	1.0125
1.0502	.0435	.8622	.9042	.0184	.9942	.9900
1.2514	.0455	.8131	.8336	.0989	.9647	.9503
1.6564	.0483	.7395	.6868	.2323	.9264	.8957
2.0471	.0493	.6744	.6120	.3370	.8932	.8487
2.5007	.0505	.6096	.5610	.4391	.8609	.7999
3.1598	.0505	.5316	.5207	.5590	.8229	.7383
3.7584	.0501	.4779	.5024	.6452	.7956	.6900
4.4159	.0495	.4214	.4937	.7196	.7721	.6437
5.3111	.0493	.3641	.4935	.7934	.7487	.5899
6.0776	.0486	.3248	.4996	.8371	.7349	.5505
6.8911	.0485	.2919	.5095	.8682	.7250	.5144
7.9276	.0483	.2553	.5252	.8921	.7174	.4740
9.6743	.0500	.2112	.5577	.9046	.7134	.4191
11.4831	.0521	.1790	.5823	.8972	.7158	.3741
13.3339	.0549	.1553	.6035	.8805	.7212	.3372
15.2123	.0583	.1376	.6311	.8597	.7277	.3064
17.1110	.0622	.1243	.6500	.8403	.7338	.2806
19.2667	.0668	.1130	.6659	.8217	.7397	.2560
21.2046	.0709	.1054	.6785	.8091	.7437	.2374
23.1768	.0751	.0994	.6875	.8003	.7465	.2210
25.2076	.0792	.0946	.6942	.7952	.7481	.2063
27.3313	.0832	.0907	.6990	.7936	.7486	.1929
29.5306	.0870	.0875	.7021	.7956	.7480	.1805
32.0348	.0906	.0847	.7035	.8012	.7462	.1687
34.7206	.0938	.0824	.7033	.8106	.7432	.1574
38.1133	.0963	.0802	.7015	.8256	.7385	.1451
41.5444	.0992	.0786	.5946	.8423	.7332	.1345
45.4554	.1010	.0771	.5948	.8614	.7271	.1242
49.9374	.1022	.0759	.5904	.8821	.7216	.1141
54.8730	.1030	.0749	.5861	.9022	.7142	.1048
60.2397	.1034	.0741	.5822	.9207	.7084	.0962
66.0735	.1036	.0734	.5789	.9371	.7032	.0883
72.4130	.1035	.0729	.5762	.9513	.6987	.0811
80.2323	.1034	.0724	.5739	.9646	.6944	.0738
87.7625	.1032	.0721	.5724	.9742	.6914	.0678
95.9758	.1030	.0719	.5715	.9819	.6891	.0623
104.9000	.1027	.0717	.5709	.9880	.6871	.0572
114.5982	.1025	.0715	.5708	.9927	.6855	.0526
125.1333	.1023	.0713	.5708	.9962	.6844	.0484
136.5980	.1021	.0712	.5711	.9987	.6836	.0445
149.0557	.1019	.0711	.5716	1.0005	.6831	.0409
162.6019	.1018	.0711	.5722	1.0016	.6827	.0376
179.2587	.1016	.0710	.5729	1.0024	.6825	.0342
201.8715	.1015	.0709	.5739	1.0027	.6824	.0305

MACH NO = 1.61 CONE ANGLE = 9.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.9476	.0478	.9446	1.1854	-.1792	1.0251	1.0125
1.9455	.0432	.8664	.9678	.1165	.9948	.9807
2.9141	.0457	.7993	.8122	.1223	.9612	.9414
3.9372	.0475	.7363	.6898	.2281	.9278	.8982
4.9227	.0491	.6763	.5992	.3562	.8865	.8412
5.9788	.0494	.6328	.5510	.4596	.8544	.7921
6.9361	.0491	.5971	.5113	.5786	.8197	.7317
7.9274	.0482	.5612	.4931	.6643	.7896	.6848
8.9395	.0471	.5330	.4828	.7551	.7618	.6294
9.9389	.0463	.5057	.4823	.8131	.7424	.5883
10.9627	.0456	.4794	.4887	.8673	.7253	.5417
11.9317	.0452	.4581	.4977	.8974	.7157	.5082
12.9195	.0451	.4453	.5121	.9209	.7093	.4708
13.9474	.0453	.4359	.5453	.9341	.7041	.4389
14.9986	.0473	.4334	.5816	.9183	.7091	.3969
16.3191	.0507	.4387	.6117	.8928	.7172	.3283
16.9869	.0541	.4216	.6351	.8653	.7259	.2912
17.9765	.0582	.4179	.6572	.8368	.7349	.2656
20.2115	.0624	.4094	.6734	.8138	.7422	.2466
22.0017	.0667	.4032	.6867	.7945	.7483	.2305
23.7691	.0711	.3979	.6973	.7799	.7533	.2165
25.7251	.0763	.3936	.7063	.7664	.7572	.2029
27.5498	.0804	.3903	.7124	.7593	.7595	.1917
29.4581	.0847	.3876	.7165	.7565	.7604	.1812
31.7059	.0892	.3851	.7188	.7586	.7597	.1702
33.9353	.0930	.3831	.7191	.7653	.7576	.1605
36.3797	.0964	.3814	.7176	.7764	.7541	.1511
38.3934	.0995	.3806	.7143	.7934	.7487	.1410
42.4887	.1013	.3793	.7089	.8126	.7426	.1319
45.9637	.1036	.3787	.7047	.8343	.7357	.1229
49.9559	.1048	.3659	.6989	.8578	.7283	.1141
55.0632	.1056	.3648	.6924	.8844	.7199	.1044
60.4048	.1060	.3640	.6868	.9076	.7125	.0960
66.6424	.1060	.3633	.6818	.9294	.7056	.0876
74.4786	.1057	.3628	.6772	.9502	.6990	.0790
81.8753	.1054	.3624	.6743	.9646	.6945	.0723
89.6175	.1050	.3622	.6724	.9756	.6910	.0665
99.7141	.1046	.3619	.6710	.9849	.6880	.0606
107.7053	.1043	.3618	.6703	.9913	.6860	.0558
117.5937	.1040	.3616	.6700	.9960	.6845	.0513
129.7941	.1037	.3615	.6702	.9996	.6833	.0467
142.4127	.1034	.3614	.6707	1.0018	.6827	.0427
156.8310	.1032	.3613	.6713	1.0031	.6823	.0389
172.8139	.1031	.3612	.6721	1.0037	.6821	.0354
201.7183	.1028	.3611	.6735	1.0039	.6820	.0305

MACH NO = 20.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0417	.9020	1.1954	-.0792	1.0291	1.0125
1.2437	.0471	.8542	.9692	.0158	.9955	.9810
1.6102	.0455	.7966	.8938	.1211	.9616	.9420
1.9302	.0473	.7351	.8910	.2265	.9283	.8991
2.1105	.0487	.6557	.8946	.3564	.8871	.8415
2.5629	.0489	.5925	.8437	.4579	.8549	.7937
3.2115	.0484	.5172	.8211	.5773	.8171	.7338
3.7946	.0476	.4617	.8010	.6637	.7898	.6872
4.5978	.0464	.3997	.7794	.7561	.7605	.6322
5.2804	.0455	.3566	.7778	.8156	.7416	.5916
6.1847	.0446	.3105	.7829	.8721	.7237	.5454
6.8757	.0441	.2797	.7910	.9041	.7136	.5121
7.9977	.0437	.2464	.8043	.9300	.7054	.4751
10.3555	.0442	.1892	.8424	.9467	.7001	.4035
12.7976	.0464	.1503	.8832	.9255	.7068	.3472
15.6498	.0494	.1271	.9152	.8943	.7167	.3089
17.3602	.0534	.1114	.9403	.8592	.7278	.2775
19.3642	.0576	.0997	.9643	.8290	.7374	.2550
21.4191	.0625	.0914	.9827	.8003	.7465	.2355
23.2433	.0672	.0858	.9963	.7783	.7535	.2215
25.1858	.0725	.0811	.7078	.7598	.7593	.2065
26.9917	.0774	.0777	.7158	.7479	.7631	.1950
29.0110	.0825	.0747	.7217	.7408	.7663	.1835
30.9883	.0872	.0723	.7249	.7398	.7657	.1735
33.1214	.0915	.0703	.7259	.7441	.7643	.1639
35.6491	.0957	.0684	.7247	.7548	.7609	.1538
38.2511	.0990	.0668	.7217	.7699	.7561	.1447
41.3529	.1019	.0652	.7169	.7906	.7496	.1351
44.5727	.1033	.0639	.7114	.8128	.7425	.1264
48.6018	.1055	.0626	.7046	.8394	.7341	.1169
52.7872	.1064	.0616	.6982	.8643	.7262	.1085
57.2155	.1072	.0607	.6917	.8903	.7180	.0997
63.4322	.1071	.0597	.6851	.9133	.7117	.0917
70.5111	.1073	.0594	.6807	.9365	.7034	.0832
77.6397	.1066	.0591	.6768	.9542	.6977	.0760
84.9167	.1062	.0587	.6740	.9678	.6934	.0699
97.1133	.1059	.0585	.6719	.9791	.6899	.0641
101.1370	.1054	.0583	.6707	.9871	.6873	.0593
110.4787	.1050	.0582	.6699	.9935	.6853	.0545
119.8685	.1046	.0581	.6697	.9977	.6840	.0504
131.0716	.1043	.0580	.6699	1.0008	.6832	.0463
142.5777	.1041	.0579	.6704	1.0025	.6824	.0427
156.5993	.1039	.0578	.6711	1.0036	.6821	.0390
171.3261	.1038	.0577	.6719	1.0040	.6820	.0357
201.9377	.1036	.0576	.6734	1.0040	.6819	.0306

MACH NO = 25.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0457	.9210	1.1954	-.0792	1.0251	1.0125
1.0429	.0471	.8534	.9648	.0154	.9951	.9811
1.3082	.0454	.7960	.8145	.1205	.9618	.9423
1.6279	.0472	.7347	.6915	.2257	.9285	.8995
2.1051	.0485	.6555	.5988	.3557	.8873	.8421
2.5546	.0488	.5925	.5495	.4571	.8552	.7945
3.2001	.0482	.5174	.5105	.5767	.8173	.7348
3.7734	.0473	.4629	.4839	.6635	.7898	.6884
4.6726	.0461	.4012	.4777	.7566	.7603	.6336
5.2533	.0451	.3571	.4756	.8169	.7412	.5931
6.1487	.0441	.3110	.4831	.8744	.7256	.5471
6.8912	.0435	.2798	.4877	.9073	.7126	.5140
7.8411	.0431	.2470	.5014	.9344	.7040	.4771
10.5485	.0425	.1825	.5441	.9524	.6983	.3961
13.2064	.0419	.1441	.5864	.9271	.7063	.3395
16.7188	.0402	.1213	.6212	.8903	.7197	.2891
18.8466	.0394	.1140	.6489	.8533	.7297	.2594
20.1999	.0381	.1045	.6712	.8193	.7415	.2467
22.2186	.0371	.0972	.6891	.7898	.7498	.2287
24.1534	.0364	.0918	.7032	.7659	.7574	.2137
25.1573	.0358	.0877	.7140	.7480	.7631	.2008
27.8459	.0383	.0747	.7213	.7370	.7665	.1900
29.8515	.0841	.0721	.7265	.7313	.7693	.1791
31.9993	.0891	.0697	.7296	.7321	.7681	.1689
34.3251	.0936	.0677	.7296	.7394	.7658	.1590
36.9053	.0976	.0659	.7266	.7531	.7615	.1493
39.7139	.1013	.0644	.7224	.7717	.7556	.1400
42.9433	.1033	.0629	.7168	.7942	.7484	.1309
46.4217	.1052	.0616	.7112	.8197	.7404	.1219
50.4544	.1064	.0605	.7033	.8461	.7328	.1131
54.9174	.1072	.0595	.6957	.8717	.7239	.1047
60.0223	.1076	.0587	.6886	.8963	.7161	.0965
66.0608	.1077	.0581	.6848	.9198	.7096	.0884
72.9544	.1074	.0575	.6789	.9409	.7020	.0806
80.4039	.1070	.0571	.6761	.9583	.6964	.0736
88.0037	.1066	.0569	.6734	.9716	.6922	.0676
95.9027	.1061	.0567	.6715	.9819	.6891	.0623
104.2555	.1056	.0566	.6703	.9886	.6865	.0576
112.6435	.1053	.0565	.6697	.9949	.6848	.0535
122.2052	.1049	.0564	.6676	.9949	.6836	.0495
132.7751	.1047	.0563	.6639	1.0016	.6827	.0457
144.4512	.1044	.0562	.6714	1.0031	.6823	.0421
157.4571	.1043	.0561	.6711	1.0038	.6820	.0388
172.1733	.1041	.0560	.6721	1.0041	.6819	.0356
201.5698	.1040	.0559	.6735	1.0039	.6821	.0327

MACH NO = 30.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0407	.9003	1.1854	-.0792	1.0251	1.0125
1.2425	.0430	.8528	.9702	.0153	.9952	.9812
1.3071	.0454	.7956	.8049	.1202	.9619	.9424
1.6252	.0472	.7344	.6918	.2253	.9286	.8997
2.1021	.0485	.6553	.5989	.3551	.8875	.8424
2.5495	.0487	.5925	.5497	.4564	.8554	.7950
3.1919	.0481	.5176	.5033	.5760	.8175	.7355
3.7692	.0472	.4623	.4835	.6630	.7900	.6892
4.5670	.0459	.4007	.4768	.7564	.7604	.6346
5.2337	.0448	.3576	.4744	.8172	.7411	.5942
6.1237	.0439	.3216	.4785	.8754	.7227	.5483
6.8696	.0433	.2894	.4858	.9089	.7121	.5153
7.8032	.0428	.2476	.4982	.9368	.7033	.4785
10.6763	.0432	.1796	.5444	.9557	.6973	.3929
13.4744	.0455	.1414	.5887	.9274	.7062	.3346
16.0878	.0491	.1157	.6244	.8881	.7147	.2939
18.4775	.0535	.1017	.6526	.8490	.7311	.2645
21.6686	.0584	.0918	.6752	.8133	.7424	.2423
23.7127	.0637	.0848	.6932	.7828	.7526	.2246
24.6692	.0692	.0797	.7072	.7584	.7598	.2100
26.5982	.0748	.0758	.7178	.7408	.7653	.1974
28.4290	.0803	.0729	.7247	.7304	.7686	.1867
30.4888	.0856	.0704	.7294	.7259	.7701	.1760
32.6984	.0904	.0682	.7312	.7285	.7692	.1658
35.1106	.0950	.0663	.7313	.7380	.7662	.1559
37.7632	.0988	.0645	.7271	.7538	.7612	.1463
40.6567	.1018	.0630	.7222	.7742	.7547	.1371
43.8971	.1041	.0616	.7161	.7983	.7471	.1281
47.6197	.1053	.0603	.7081	.8249	.7387	.1191
51.6949	.1059	.0592	.7022	.8512	.7304	.1106
56.2323	.1076	.0583	.6958	.8764	.7224	.1025
61.4675	.1079	.0575	.6897	.9035	.7147	.0944
67.6446	.1081	.0569	.6841	.9235	.7075	.0864
74.5216	.1077	.0565	.6795	.9437	.7011	.0790
81.9580	.1073	.0562	.6757	.9605	.6958	.0723
89.8399	.1068	.0559	.6771	.9734	.6917	.0665
97.4096	.1063	.0558	.6713	.9833	.6895	.0614
105.7160	.1058	.0557	.6701	.9908	.6862	.0568
113.9885	.1054	.0556	.6696	.9958	.6846	.0529
123.5463	.1051	.0555	.6695	.9996	.6834	.0490
133.9752	.1048	.0554	.6698	1.0020	.6826	.0453
145.4491	.1046	.0553	.6704	1.0034	.6822	.0419
158.1659	.1045	.0552	.6711	1.0039	.6820	.0386
172.3576	.1044	.0551	.6720	1.0040	.6821	.0355
200.1925	.1042	.0550	.6735	1.0036	.6821	.0327

MACH NO = 3.50 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0418	.9937	1.2101	-.0882	1.0311	1.0154
.9712	.1447	.9798	1.0382	-.0184	1.0065	.9898
1.1434	.0478	.9154	.9660	.0516	.9818	.9609
1.4210	.0522	.8538	.7731	.1440	.9492	.9177
1.7603	.0565	.7876	.6945	.2339	.9175	.8699
2.1671	.0613	.7202	.6389	.3204	.8870	.8189
2.6458	.0633	.6579	.6019	.4034	.8577	.7659
3.1904	.0659	.5916	.5805	.4783	.8313	.7126
3.8300	.0683	.5333	.5705	.5431	.8085	.6603
4.5387	.0704	.4816	.5642	.5978	.7892	.6103
5.3263	.0724	.4363	.5619	.6427	.7734	.5623
6.1935	.0743	.3973	.5770	.6783	.7618	.5178
7.1408	.0763	.3642	.5850	.7061	.7515	.4766
8.7131	.0792	.3243	.5979	.7374	.7410	.4210
10.1638	.0814	.2986	.6192	.7567	.7351	.3800
11.7443	.0835	.2791	.6174	.7721	.7277	.3437
13.4571	.0853	.2620	.6254	.7847	.7233	.3113
15.6920	.0873	.2472	.6335	.7975	.7188	.2773
17.7093	.0887	.2377	.6391	.8067	.7155	.2524
19.8734	.0900	.2312	.6478	.8148	.7126	.2302
22.1918	.0911	.2242	.6479	.8221	.7101	.2104
24.6734	.0921	.2194	.6514	.8287	.7078	.1927
27.8826	.0931	.2149	.6550	.8358	.7058	.1737
30.7661	.0937	.2120	.6575	.8410	.7034	.1596
33.8576	.0943	.2096	.6598	.8458	.7017	.1469
37.1768	.0948	.2076	.6618	.8502	.7002	.1352
40.7458	.0953	.2060	.6636	.8543	.6987	.1246
45.3926	.0957	.2045	.6655	.8586	.6972	.1131
49.5987	.0960	.2034	.6669	.8620	.6961	.1043
54.1374	.0963	.2026	.6681	.8650	.6951	.0963
59.0791	.0965	.2019	.6693	.8678	.6941	.0889
65.4404	.0967	.2012	.6706	.8707	.6929	.0818
71.2497	.0968	.2007	.6716	.8729	.6922	.0746
77.5295	.0969	.2003	.6726	.8748	.6915	.0689
84.7161	.0970	.2000	.6735	.8765	.6909	.0637
91.6547	.0971	.1997	.6743	.8780	.6904	.0588
101.2534	.0971	.1995	.6753	.8795	.6898	.0535
109.9707	.0972	.1993	.6760	.8806	.6895	.0494
119.3930	.0972	.1992	.6767	.8815	.6891	.0457
123.5966	.0972	.1991	.6774	.8823	.6888	.0422
140.6258	.0972	.1989	.6781	.8830	.6886	.0390
155.0591	.0973	.1988	.6788	.8837	.6884	.0355
169.1702	.0973	.1988	.6794	.8841	.6882	.0328
182.3534	.0973	.1987	.6799	.8845	.6881	.0303
200.0128	.0973	.1987	.6805	.8849	.6879	.0276

MACH NO = [illegible] CONE ANGLE = 10.0 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0412	.9827	1.2111	-.1882	1.0311	1.154
.9766	.0436	.9130	1.0323	-.1156	1.0035	.9388
1.1655	.0463	.8652	.8833	.0617	.9792	.9573
1.4747	.0497	.7967	.7541	.1649	.9418	.8999
1.8404	.0526	.7276	.6677	.2627	.9073	.8594
2.2667	.0549	.6591	.6124	.3547	.8749	.8272
2.7605	.0564	.5914	.5749	.4405	.9447	.7542
3.3237	.0576	.5249	.5576	.5166	.8178	.7217
3.9547	.0586	.4719	.5483	.5813	.7956	.6959
4.6572	.0597	.4212	.5492	.6334	.7766	.6726
5.4163	.0603	.3769	.5541	.6733	.7627	.5574
6.2418	.0623	.3398	.5640	.7016	.7526	.5155
7.1277	.0633	.3063	.5759	.7212	.7457	.4771
8.5674	.0668	.2668	.5909	.7385	.7396	.4256
10.1416	.0701	.2363	.6131	.7467	.7367	.3886
11.8543	.0735	.2128	.6289	.7509	.7352	.3414
13.7192	.0771	.1948	.6418	.7541	.7341	.3169
15.7483	.0815	.1809	.6518	.7583	.7326	.2766
17.9634	.0836	.1702	.6592	.7641	.7315	.2496
20.3434	.0864	.1621	.6644	.7713	.7299	.2259
22.8521	.0888	.1558	.6679	.7794	.7281	.2054
25.5008	.0909	.1511	.6713	.7879	.7261	.1874
28.3096	.0926	.1475	.6719	.7966	.7191	.1715
31.3006	.0941	.1446	.6729	.8051	.7161	.1573
34.5044	.0952	.1423	.6735	.8135	.7131	.1444
37.9477	.0962	.1405	.6739	.8218	.7102	.1328
41.6556	.0969	.1391	.6740	.8297	.7074	.1222
45.6566	.0975	.1379	.6740	.8372	.7047	.1125
49.9789	.0981	.1369	.6739	.8443	.7023	.1036
54.6527	.0983	.1362	.6739	.8507	.7000	.0955
59.7104	.0985	.1355	.6739	.8565	.6979	.0880
65.1864	.0987	.1350	.6739	.8617	.6961	.0811
71.1173	.0988	.1346	.6741	.8663	.6945	.0747
77.5425	.0988	.1342	.6743	.8702	.6931	.0689
84.5042	.0989	.1340	.6746	.8736	.6919	.0635
92.0480	.0989	.1337	.6749	.8765	.6909	.0586
100.2231	.0988	.1335	.6754	.8789	.6901	.0540
109.0831	.0988	.1334	.6758	.8808	.6894	.0498
118.6857	.0987	.1332	.6763	.8824	.6888	.0459
129.0937	.0987	.1331	.6769	.8836	.6884	.0424
140.3749	.0986	.1330	.6775	.8846	.6880	.0391
152.6033	.0986	.1330	.6781	.8853	.6878	.0360
165.8579	.0985	.1329	.6786	.8859	.6876	.0332
180.2260	.0985	.1328	.6792	.8863	.6875	.0307
201.2785	.0985	.1328	.6799	.8866	.6873	.0275

MACH NO = 10.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0406	.9162	1.2131	-.3882	1.0311	1.0154
1.0230	.0432	.8641	.9895	.3053	.9981	.9809
1.2104	.0451	.8188	.9603	.1803	.9717	.9511
1.5836	.0477	.7394	.7137	.2027	.9285	.8942
1.9397	.0491	.6744	.6382	.2971	.8962	.8467
2.4596	.0499	.5948	.5765	.4092	.8557	.7857
2.9375	.0500	.5342	.5453	.4922	.8264	.7369
3.6079	.0497	.4648	.5228	.5836	.7942	.6778
4.1972	.0493	.4153	.5151	.6444	.7728	.6332
4.9903	.0489	.3616	.5154	.7035	.7519	.5817
5.6619	.0488	.3249	.5212	.7377	.7398	.5442
6.3596	.0490	.2933	.5314	.7617	.7314	.5111
7.2600	.0495	.2600	.5449	.7796	.7251	.4719
8.9339	.0513	.2144	.5743	.7876	.7223	.4142
10.6469	.0541	.1821	.6032	.7776	.7250	.3681
12.1853	.0572	.1613	.6282	.7623	.7312	.3347
13.9213	.0613	.1439	.6480	.7438	.7377	.3036
15.6553	.0657	.1310	.6656	.7272	.7436	.2778
17.3922	.0702	.1214	.6734	.7138	.7483	.2560
18.9460	.0743	.1148	.6889	.7050	.7514	.2392
20.7267	.0788	.1090	.6971	.6985	.7537	.2225
22.5726	.0832	.1043	.7029	.6957	.7547	.2075
24.5229	.0874	.1006	.7067	.6966	.7543	.1937
26.6234	.0913	.0975	.7085	.7013	.7527	.1807
28.6593	.0944	.0952	.7085	.7089	.7500	.1697
31.1964	.0974	.0930	.7069	.7211	.7457	.1577
34.0633	.0999	.0911	.7038	.7369	.7401	.1461
37.3390	.1018	.0895	.6996	.7556	.7335	.1347
41.1297	.1031	.0881	.6946	.7763	.7262	.1236
45.0298	.1037	.0870	.6899	.7954	.7195	.1139
49.9857	.1040	.0861	.6849	.8158	.7123	.1036
55.4234	.1039	.0853	.6808	.8337	.7060	.0942
61.3856	.1037	.0848	.6776	.8485	.7008	.0857
67.1649	.1034	.0844	.6755	.8593	.6970	.0789
74.2555	.1031	.0840	.6740	.8688	.6936	.0718
82.0294	.1027	.0837	.6732	.8760	.6911	.0653
90.5514	.1024	.0835	.6729	.8812	.6892	.0595
99.8993	.1021	.0834	.6731	.8848	.6880	.0542
108.9673	.1019	.0832	.6735	.8870	.6872	.0499
120.1038	.1017	.0831	.6743	.8885	.6867	.0454
132.3246	.1015	.0830	.6751	.8894	.6864	.0414
145.7364	.1014	.0829	.6760	.8898	.6862	.0377
160.4564	.1013	.0829	.6769	.8899	.6862	.0343
174.7425	.1012	.0828	.6776	.8899	.6862	.0316
200.6348	.1011	.0828	.6788	.8896	.6863	.0276

MACH NO = 15.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0415	.9091	1.2171	-.0882	1.0311	1.0154
1.0185	.0429	.8582	.9932	.1034	.9988	.9817
1.2684	.0452	.7988	.8292	.1019	.9641	.9410
1.5658	.0470	.7361	.7169	.1997	.9299	.8967
2.0073	.0484	.6564	.6246	.3158	.8886	.8382
2.4180	.0488	.5939	.5758	.4053	.8571	.7902
3.0046	.0486	.5198	.5363	.5089	.8215	.7335
3.5279	.0480	.4655	.5174	.5826	.7945	.6844
4.2410	.0472	.4051	.5160	.6601	.7672	.6302
4.8508	.0466	.3671	.5046	.7092	.7499	.5902
5.6505	.0461	.3182	.5037	.7548	.7378	.5448
6.3124	.0459	.2879	.5176	.7799	.7291	.5123
7.1578	.0460	.2559	.5316	.7995	.7191	.4759
8.2227	.0475	.2230	.5675	.8091	.7147	.4256
11.1071	.0501	.1667	.6017	.7927	.7214	.3575
13.1071	.0540	.1423	.6316	.7654	.7297	.3174
14.8676	.0581	.1269	.6546	.7420	.7383	.2890
16.7065	.0623	.1151	.6746	.7182	.7467	.2642
18.3330	.0675	.1072	.6892	.6997	.7532	.2456
20.0727	.0729	.1018	.7017	.6838	.7589	.2284
21.6681	.0778	.0982	.7114	.6733	.7636	.2146
23.4496	.0830	.0953	.7172	.6666	.7649	.2010
25.1599	.0876	.0933	.7211	.6651	.7655	.1896
27.1612	.0922	.0866	.7229	.6686	.7642	.1777
29.1660	.0961	.0845	.7225	.6766	.7614	.1672
31.6881	.0997	.0825	.7189	.6904	.7565	.1560
34.0799	.1023	.0809	.7158	.7072	.7506	.1460
37.1235	.1044	.0794	.7130	.7290	.7429	.1354
40.3296	.1057	.0782	.7098	.7512	.7351	.1258
44.3091	.1064	.0770	.6968	.7765	.7262	.1156
48.4728	.1066	.0761	.6936	.7990	.7182	.1065
53.7862	.1064	.0754	.6844	.8223	.7101	.0969
59.3392	.1060	.0748	.6797	.8408	.7035	.0885
65.6179	.1054	.0744	.6762	.8562	.6981	.0806
71.5725	.1050	.0741	.6740	.8667	.6944	.0743
78.4243	.1045	.0739	.6726	.8752	.6914	.0682
85.1666	.1041	.0737	.6720	.8808	.6894	.0631
93.1970	.1037	.0736	.6719	.8851	.6879	.0579
101.3538	.1034	.0734	.6722	.8878	.6869	.0534
111.3991	.1031	.0733	.6729	.8896	.6863	.0488
121.9465	.1030	.0732	.6737	.8904	.6860	.0448
135.3878	.1028	.0731	.6748	.8909	.6858	.0405
150.0190	.1026	.0730	.6759	.8909	.6858	.0366
168.4020	.1025	.0730	.6770	.8907	.6859	.0328
200.1422	.1023	.0729	.6736	.8902	.6861	.0277

MACH NO = 20.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0433	.9365	1.2101	-.0882	1.0311	1.0154
1.0168	.0429	.8562	.3945	.0027	.9990	.9819
1.2645	.0459	.7972	.3318	.1007	.9645	.9416
1.5592	.0468	.7349	.7181	.1972	.9304	.8976
1.9966	.0481	.6557	.6251	.3142	.8892	.8395
2.5120	.0484	.5785	.5661	.4251	.8511	.7900
2.9809	.0481	.5200	.5354	.5074	.8211	.7328
3.6318	.0473	.4533	.5121	.5988	.7888	.6759
4.3443	.0463	.3950	.5018	.6741	.7623	.6230
4.9496	.0457	.3545	.5018	.7217	.7455	.5842
5.7390	.0451	.3113	.5060	.7658	.7299	.5402
6.3888	.0449	.2821	.5137	.7902	.7213	.5187
7.2148	.0448	.2513	.5263	.8093	.7146	.4736
9.5471	.0452	.1935	.5682	.8171	.7118	.3964
11.8195	.0454	.1539	.6080	.7926	.7205	.3421
13.9653	.0457	.1309	.6401	.7612	.7316	.3029
15.9667	.0586	.1160	.6653	.7311	.7422	.2736
17.8393	.0639	.1059	.6852	.7047	.7515	.2510
19.6000	.0691	.0981	.6998	.6843	.7587	.2338
21.2493	.0748	.0937	.7119	.6675	.7646	.2181
23.0068	.0805	.0895	.7206	.6566	.7685	.2043
24.8261	.0861	.0862	.7261	.6517	.7702	.1917
26.7682	.0913	.0835	.7298	.6530	.7697	.1799
28.8804	.0960	.0812	.7297	.6605	.7671	.1686
31.2239	.0999	.0792	.7266	.6742	.7622	.1576
33.8111	.1031	.0775	.7213	.6929	.7557	.1470
36.4858	.1052	.0761	.7155	.7137	.7483	.1375
39.7752	.1067	.0747	.7082	.7389	.7394	.1273
43.4397	.1075	.0735	.7008	.7645	.7304	.1177
47.5857	.1077	.0726	.6937	.7895	.7216	.1083
52.4248	.1076	.0718	.6871	.8134	.7132	.0992
57.8911	.1071	.0713	.6816	.8343	.7058	.0905
63.8616	.1065	.0709	.6774	.8514	.6998	.0826
69.9315	.1059	.0706	.6745	.8641	.6953	.0759
75.7443	.1054	.0704	.6729	.8728	.6922	.0704
82.7152	.1049	.0702	.6718	.8797	.6898	.0651
89.3357	.1045	.0701	.6714	.8846	.6881	.0603
96.9459	.1041	.0700	.6716	.8877	.6869	.0558
105.3048	.1038	.0698	.6721	.8897	.6863	.0515
114.6045	.1037	.0697	.6729	.8907	.6859	.0475
125.0876	.1035	.0696	.6738	.8910	.6858	.0437
137.0734	.1034	.0695	.6749	.8910	.6858	.0400
149.9267	.1033	.0695	.6759	.8909	.6858	.0367
166.1864	.1032	.0694	.6769	.8906	.6859	.0332
201.6595	.1030	.0693	.6786	.8901	.6861	.0276

MACH NO = 25.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID CA	AERODYNAMIC XCP/L	COEFFICIENTS YCP/D	XVCP/LV	RN/RB
.9264	.0495	.9055	1.2131	-.0882	1.0311	1.0154
1.0161	.0427	.8553	.9952	.1024	.9991	.9821
1.2628	.0449	.7966	.9315	.1002	.9647	.9418
1.5562	.0467	.7345	.7187	.1966	.9307	.8981
1.9916	.0480	.6556	.6253	.3135	.8895	.8401
2.5044	.0482	.5785	.5659	.4244	.8513	.7868
2.9787	.0479	.5202	.5349	.5069	.8212	.7337
3.6173	.0470	.4536	.5111	.5987	.7889	.6771
4.3244	.0460	.3953	.5003	.6746	.7621	.6244
4.9246	.0453	.3549	.4948	.7228	.7451	.5857
5.7063	.0446	.3118	.4934	.7678	.7292	.5419
6.3490	.0443	.2826	.5138	.7929	.7204	.5106
7.1650	.0442	.2518	.5229	.8129	.7133	.4756
9.7860	.0457	.1845	.5730	.8234	.7117	.3899
12.1473	.0491	.1484	.6110	.7925	.7215	.3355
14.4822	.0538	.1251	.6454	.7569	.7331	.2948
16.4879	.0588	.1113	.6734	.7253	.7442	.2669
18.4698	.0647	.1014	.6913	.6960	.7545	.2442
20.2340	.0705	.0950	.7064	.6739	.7623	.2269
22.0773	.0768	.0899	.7184	.6569	.7683	.2113
23.8265	.0826	.0862	.7260	.6475	.7716	.1984
25.7830	.0885	.0831	.7307	.6446	.7727	.1857
27.7604	.0937	.0817	.7321	.6485	.7713	.1744
30.0745	.0984	.0785	.7335	.6598	.7673	.1628
32.4671	.1020	.0767	.7265	.6764	.7615	.1523
35.2783	.1048	.0750	.7214	.6987	.7536	.1417
38.2822	.1066	.0736	.7133	.7231	.7450	.1318
41.8421	.1078	.0723	.7054	.7502	.7355	.1217
45.5453	.1082	.0713	.6982	.7749	.7267	.1127
50.0960	.1083	.0704	.6929	.8004	.7177	.1034
54.9759	.1079	.0698	.6850	.8222	.7111	.0949
60.7339	.1074	.0693	.6798	.8420	.7030	.0866
66.5073	.1067	.0690	.6762	.8570	.6978	.0796
72.7413	.1061	.0688	.6736	.8687	.6936	.0732
78.7709	.1055	.0686	.6721	.8767	.6918	.0679
85.5577	.1050	.0685	.6713	.8828	.6887	.0628
92.3461	.1046	.0684	.6712	.8867	.6873	.0584
100.1857	.1043	.0683	.6715	.8893	.6864	.0540
108.7977	.1041	.0681	.6722	.8906	.6859	.0499
117.7342	.1039	.0680	.6730	.8911	.6857	.0463
128.3946	.1038	.0679	.6741	.8911	.6857	.0426
139.6883	.1037	.0678	.6751	.8909	.6858	.0393
153.4745	.1036	.0678	.6762	.8905	.6860	.0358
168.4824	.1035	.0677	.6772	.8902	.6861	.0327
201.2558	.1034	.0676	.6797	.8900	.6862	.0275

MACH NO = 30.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 3.0.

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0435	.9048	1.2101	-.1882	1.0311	1..154
1.0156	.0427	.8548	.3955	.0022	.9992	.9821
1.2618	.0449	.7961	.8319	.0999	.9648	.9420
1.5546	.0465	.7342	.7190	.1962	.9308	.8983
1.9888	.0479	.6554	.6255	.3130	.8896	.8465
2.5002	.0481	.5794	.5659	.4240	.8515	.7813
2.9650	.0479	.5202	.5346	.5066	.8214	.7343
3.6093	.0469	.4537	.5136	.5986	.7889	.6777
4.3135	.0459	.3955	.4995	.6748	.7620	.6251
4.9108	.0451	.3552	.4977	.7233	.7449	.5865
5.6884	.0444	.3126	.5020	.7689	.7289	.5429
6.3271	.0440	.2829	.5032	.7944	.7199	.5116
7.1375	.0439	.2521	.5211	.8148	.7126	.4767
9.8958	.0454	.1817	.5716	.8223	.7100	.3870
12.3676	.0489	.1459	.6133	.7919	.7207	.3311
14.7729	.0539	.1220	.6483	.7542	.7340	.2904
16.8258	.0591	.1085	.6738	.7211	.7457	.2627
18.8346	.0653	.0990	.6949	.6906	.7564	.2404
20.7368	.0717	.0925	.7109	.6669	.7648	.2224
22.4945	.0778	.0880	.7217	.6513	.7703	.2081
24.3974	.0842	.0843	.7293	.6423	.7735	.1945
26.2927	.0899	.0815	.7331	.6410	.7749	.1826
28.4690	.0953	.0791	.7336	.6472	.7718	.1707
30.7324	.0997	.0771	.7311	.6601	.7672	.1598
33.3503	.1033	.0752	.7261	.6798	.7603	.1488
36.2789	.1058	.0736	.7191	.7041	.7517	.1382
39.4128	.1074	.0722	.7116	.7297	.7427	.1284
43.0398	.1083	.0710	.7036	.7568	.7331	.1186
46.8653	.1086	.0700	.6965	.7814	.7244	.1099
51.5855	.1085	.0692	.6894	.8064	.7156	.1007
56.5167	.1081	.0687	.6837	.8272	.7083	.0926
62.3176	.1075	.0683	.6789	.8460	.7016	.0845
68.4404	.1068	.0680	.6753	.8608	.6964	.0775
74.2764	.1061	.0678	.6731	.8711	.6928	.0718
80.6894	.1056	.0677	.6717	.8790	.6900	.0664
87.0421	.1051	.0676	.6711	.8842	.6882	.0618
94.2541	.1047	.0674	.6711	.8879	.6869	.0573
102.0760	.1044	.0673	.6715	.8901	.6861	.0531
110.1254	.1042	.0672	.6722	.8910	.6858	.0494
119.5645	.1041	.0671	.6732	.8913	.6857	.0456
129.4477	.1040	.0670	.6742	.8911	.6858	.0423
141.2325	.1039	.0669	.6753	.8907	.6859	.0388
153.7997	.1038	.0668	.6763	.8902	.6861	.0358
169.1223	.1038	.0668	.6773	.8899	.6862	.0326
200.5079	.1036	.0667	.6788	.8897	.6862	.0276

NSWC/WOL/TR 75-45

MACH NO = 3.50 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0401	1.0298	1.3492	-.1340	1.0718	1.0353
.9149	.0439	.9611	1.1102	-.1498	1.0257	.9877
1.0720	.0467	.9171	.9812	-.1101	.9946	.9483
1.3079	.0505	.8571	.9618	-.0809	.9567	.9946
1.5728	.0540	.7718	.7849	.1423	.9237	.8412
1.9675	.0579	.7094	.7342	.1957	.8951	.7889
2.2772	.0602	.6406	.6938	.2531	.8644	.7259
2.6459	.0623	.5898	.6777	.2924	.8433	.6774
3.0490	.0644	.5459	.6621	.3251	.8258	.6312
3.4861	.0664	.5056	.6561	.3523	.8112	.5877
3.9559	.0684	.4715	.6547	.3737	.7997	.5473
4.4573	.0704	.4422	.6562	.3906	.7917	.5098
5.1275	.0723	.4116	.6601	.4070	.7819	.4670
5.9951	.0759	.3824	.6661	.4216	.7741	.4213
7.0964	.0792	.3566	.6773	.4343	.7673	.3747
8.3019	.0823	.3375	.6737	.4445	.7618	.3342
9.4296	.0847	.3251	.6841	.4522	.7577	.3036
10.8751	.0871	.3141	.6880	.4609	.7531	.2716
12.4874	.0892	.3058	.6918	.4693	.7485	.2431
14.2863	.0908	.2996	.6927	.4778	.7443	.2176
15.9584	.0919	.2957	.6937	.4846	.7413	.1983
18.0778	.0929	.2922	.6944	.4922	.7362	.1782
20.4026	.0936	.2897	.6949	.4992	.7325	.1604
23.5796	.0940	.2881	.6952	.5047	.7295	.1467
25.3574	.0943	.2866	.6956	.5104	.7265	.1322
28.4227	.0945	.2855	.6960	.5154	.7238	.1193
31.3334	.0945	.2847	.6964	.5190	.7219	.1092
34.9913	.0946	.2841	.6971	.5225	.7201	.0986
39.0697	.0945	.2836	.6978	.5253	.7185	.0890
43.5785	.0945	.2832	.6987	.5275	.7173	.0804
47.8237	.0944	.2829	.6995	.5290	.7165	.0736
53.2649	.0944	.2827	.7005	.5304	.7158	.0665
59.2862	.0943	.2825	.7015	.5313	.7153	.0600
64.0567	.0943	.2824	.7024	.5320	.7149	.0550
72.2256	.0942	.2823	.7033	.5325	.7147	.0497
80.2716	.0942	.2822	.7043	.5328	.7145	.0449
87.8497	.0941	.2821	.7051	.5330	.7144	.0411
97.5654	.0941	.2821	.7059	.5331	.7143	.0372
109.3205	.0941	.2820	.7068	.5331	.7143	.0336
118.4513	.0940	.2820	.7074	.5331	.7143	.0308
131.4394	.0940	.2820	.7081	.5331	.7143	.0278
145.8181	.0940	.2820	.7087	.5330	.7144	.0251
161.7354	.0940	.2819	.7093	.5330	.7144	.0227
176.7276	.0940	.2819	.7098	.5329	.7144	.0208
191.8140	.0940	.2819	.7104	.5329	.7144	.0182

MACH NO = 5.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0397	.9825	1.3432	-.1340	1.0718	1..353
.9035	.0426	.9194	1.1239	-..541	1.0229	.93.7
1.0964	.0454	.8521	.3643	.1205	.9896	.9424
1.3153	.0480	.7870	.3514	.1873	.9532	.8330
1.6262	.0507	.7094	.7620	.1608	.9128	.8312
1.9039	.0524	.6515	.7155	.2120	.8864	.7828
2.2864	.0543	.5855	.6777	.2670	.8569	.7247
2.6231	.0555	.5376	.6590	.3042	.8370	.6802
3.0794	.0571	.4847	.6469	.3414	.8171	.6281
3.4670	.0584	.4496	.6436	.3642	.8048	.5835
3.9787	.0603	.4086	.6453	.3849	.7977	.5454
4.4059	.0619	.3818	.6493	.3966	.7875	.5134
4.9596	.0641	.3576	.6563	.4066	.7821	.4771
5.8804	.0680	.3183	.6732	.4151	.7776	.4268
6.9454	.0721	.2924	.6816	.4189	.7755	.3844
7.8565	.0753	.2731	.6920	.4212	.7743	.3481
9.0581	.0808	.2573	.7008	.4243	.7726	.3130
10.1951	.0845	.2468	.7059	.4284	.77 4	.2858
11.4148	.0878	.2389	.7090	.4341	.7674	.2614
12.7410	.0905	.2328	.7103	.4414	.7655	.2392
14.2047	.0929	.2281	.7103	.4503	.7589	.2187
15.8402	.0948	.2245	.7091	.4598	.7556	.1995
17.9420	.0962	.2213	.7067	.4720	.7471	.1794
20.0997	.0973	.2191	.7041	.4833	.7410	.1625
22.5614	.0974	.2175	.7015	.4943	.7351	.1468
25.2802	.0975	.2162	.6993	.5040	.7299	.1326
29.2770	.0973	.2153	.6977	.5122	.7255	.1198
31.5830	.0971	.2146	.6968	.5187	.7221	.1083
35.2319	.0969	.2140	.6965	.5237	.7193	.0980
39.7921	.0967	.2135	.6968	.5277	.7172	.0875
44.2954	.0965	.2132	.6975	.5302	.7159	.0791
49.2688	.0963	.2129	.6984	.5318	.7150	.0716
54.7619	.0961	.2127	.6995	.5329	.7144	.0648
60.8297	.0960	.2126	.7006	.5335	.7141	.0586
67.5328	.0959	.2124	.7018	.5339	.7139	.0530
75.9169	.0959	.2123	.7030	.5341	.7138	.0474
84.2015	.0958	.2123	.7041	.5341	.7138	.0429
93.7552	.0957	.2122	.7051	.5340	.7138	.0388
103.4695	.0957	.2122	.7060	.5339	.7139	.0351
114.6457	.0957	.2121	.7068	.5337	.7141	.0318
126.9955	.0956	.2121	.7076	.5336	.7141	.0287
140.6423	.0956	.2121	.7083	.5334	.7141	.0260
157.7156	.0956	.2121	.7090	.5333	.7142	.0232
174.5894	.0956	.2120	.7096	.5332	.7143	.0210
200.7226	.0956	.2120	.7103	.5330	.7143	.0183

MACH NO = 10.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0394	.9456	1.3492	-.1340	1.0718	1.0353
.9289	.0421	.8727	1.0933	-.0426	1.0228	.9840
1.1083	.0440	.8118	.9491	.0265	.9858	.9396
1.3654	.0460	.7364	.8253	.1052	.9476	.8825
1.6584	.0475	.6643	.7438	.1753	.9060	.8253
1.9848	.0485	.5974	.6934	.2365	.8733	.7697
2.4166	.0492	.5255	.6518	.2982	.8402	.7068
2.8065	.0497	.4730	.6325	.3394	.8181	.6582
3.2165	.0502	.4276	.6242	.3710	.8012	.6138
3.6402	.0510	.3889	.6230	.3937	.7890	.5738
4.0718	.0519	.3562	.6268	.4088	.7819	.5381
4.5068	.0532	.3287	.6338	.4178	.7761	.5064
5.0292	.0549	.3014	.6446	.4226	.7735	.4728
6.0643	.0594	.2611	.6687	.4199	.7750	.4180
7.0764	.0647	.2339	.6907	.4108	.7798	.3755
8.0627	.0703	.2151	.7081	.4019	.7846	.3416
9.1094	.0764	.2010	.7220	.3947	.7885	.3117
10.0709	.0818	.1917	.7311	.3907	.7916	.2885
11.0480	.0870	.1848	.7371	.3896	.7912	.2683
12.0633	.0918	.1796	.7403	.3916	.7901	.2500
13.1414	.0960	.1756	.7418	.3972	.7872	.2332
14.4128	.0997	.1723	.7385	.4072	.7818	.2160
15.7174	.1021	.1698	.7339	.4200	.7749	.2009
17.1830	.1036	.1679	.7275	.4358	.7665	.1862
18.8398	.1041	.1663	.7199	.4537	.7569	.1720
20.7229	.1039	.1652	.7119	.4724	.7468	.1582
23.0506	.1030	.1639	.7037	.4917	.7365	.1440
25.4848	.1020	.1631	.6977	.5070	.7283	.1316
28.1845	.1011	.1625	.6938	.5187	.7220	.1202
31.1198	.1002	.1620	.6918	.5266	.7178	.1098
34.3523	.0996	.1615	.6915	.5314	.7152	.1003
38.3092	.0992	.1611	.6923	.5342	.7137	.0906
42.5289	.0989	.1607	.6938	.5353	.7131	.0822
47.4836	.0988	.1604	.6957	.5356	.7130	.0741
53.4213	.0986	.1602	.6977	.5355	.7130	.0663
61.1978	.0985	.1599	.6997	.5353	.7132	.0583
69.3837	.0984	.1598	.7014	.5350	.7133	.0517
78.6115	.0983	.1597	.7029	.5348	.7134	.0458
89.0143	.0982	.1596	.7043	.5346	.7135	.0406
100.7427	.0982	.1595	.7055	.5344	.7136	.0360
115.1419	.0981	.1595	.7067	.5341	.7138	.0316
130.2025	.0981	.1595	.7076	.5338	.7139	.0280
147.1850	.0981	.1594	.7085	.5336	.7141	.0249
166.3349	.0981	.1594	.7093	.5334	.7142	.0221
201.7767	.0981	.1594	.7103	.5331	.7143	.0182

MACH NO = 15.00 CONE ANGLE = 15.0 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0393	.9385	1.3492	-.1340	1.0718	1.0353
.9247	.0418	.8672	1.0972	-.1443	1.0237	.9851
1.1000	.0436	.8075	.9536	.3240	.9871	.9416
1.3511	.0455	.7333	.8290	.1023	.9452	.8855
1.6980	.0469	.6483	.7330	.1855	.9006	.8181
2.0217	.0477	.5832	.6824	.2453	.8686	.7639
2.3738	.0480	.5243	.6492	.2966	.8410	.7126
2.7505	.0483	.4721	.6293	.3389	.8184	.6648
3.2253	.0487	.4186	.6182	.3773	.7978	.6129
3.6322	.0492	.3812	.6169	.3998	.7858	.5745
4.0435	.0499	.3495	.6203	.4148	.7777	.5403
4.4551	.0509	.3228	.6270	.4237	.7729	.5099
4.9458	.0524	.2962	.6376	.4283	.7705	.4779
6.1391	.0573	.2489	.6680	.4225	.7736	.4145
7.2688	.0634	.2193	.6948	.4086	.7810	.3683
8.3345	.0698	.2002	.7148	.3961	.7877	.3333
9.3514	.0763	.1874	.7296	.3863	.7930	.3155
10.4121	.0832	.1780	.7408	.3793	.7967	.2811
11.4146	.0893	.1717	.7474	.3768	.7981	.2614
12.4550	.0948	.1671	.7513	.3789	.7970	.2436
13.5649	.0995	.1636	.7497	.3856	.7933	.2272
14.7776	.1031	.1608	.7460	.3971	.7872	.2116
16.1212	.1055	.1586	.7394	.4129	.7787	.1966
17.6116	.1066	.1569	.7307	.4322	.7684	.1823
19.3890	.1066	.1554	.7202	.4549	.7562	.1677
21.2416	.1057	.1544	.7105	.4761	.7449	.1548
23.2964	.1043	.1536	.7019	.4954	.7345	.1427
25.5463	.1029	.1529	.6954	.5111	.7261	.1314
27.9749	.1017	.1524	.6913	.5225	.7200	.1210
30.5642	.1009	.1521	.6895	.5298	.7161	.1116
33.3429	.1003	.1516	.6894	.5338	.7139	.1031
36.5884	.0999	.1512	.6914	.5357	.7129	.0946
39.9752	.0997	.1510	.6921	.5362	.7127	.0871
43.7934	.0996	.1505	.6940	.5360	.7128	.0800
48.1754	.0996	.1503	.6959	.5355	.7131	.0731
53.3087	.0995	.1500	.6978	.5352	.7132	.0664
59.4624	.0994	.1498	.6995	.5350	.7133	.0599
67.5844	.0993	.1497	.7012	.5349	.7133	.0530
77.2316	.0992	.1495	.7027	.5349	.7133	.0466
89.2038	.0991	.1494	.7042	.5348	.7134	.0405
103.0216	.0990	.1494	.7056	.5345	.7136	.0352
118.9033	.0989	.1493	.7069	.5341	.7138	.0306
137.1785	.0989	.1493	.7080	.5337	.714	.0266
158.1898	.0989	.1493	.7090	.5333	.7142	.0232
200.4461	.0989	.1492	.7103	.5329	.7144	.0184

MACH NO = 20.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0393	.9359	1.3492	-.1340	1.0718	1.0353
.9232	.0417	.8651	1.0987	-.1449	1.0241	.9855
1.0970	.0435	.8058	.9553	-.1231	.9876	.9423
1.3998	.0456	.7176	.8109	.1158	.9379	.8753
1.6895	.0467	.6475	.7337	.1844	.9012	.8197
2.0100	.0473	.5826	.6825	.2443	.8691	.7658
2.4309	.0477	.5130	.6437	.3053	.8364	.7049
2.8067	.0478	.4623	.6254	.3460	.8146	.6582
3.1980	.0481	.4185	.6164	.3777	.7976	.6157
3.6788	.0486	.3743	.6147	.4043	.7833	.5704
4.0827	.0492	.3437	.6184	.4185	.7757	.5373
4.4855	.0502	.3179	.6252	.4268	.7713	.5078
4.9640	.0516	.2921	.6359	.4308	.7691	.4768
6.2691	.0570	.2413	.6703	.4223	.7737	.4086
7.4814	.0627	.2112	.6992	.4059	.7825	.3608
8.5414	.0703	.1934	.7191	.3925	.7897	.3272
9.6065	.0774	.1809	.7348	.3812	.7957	.2993
10.6446	.0845	.1724	.7457	.3739	.7996	.2763
11.6924	.0912	.1664	.7521	.3715	.8009	.2564
12.7861	.0970	.1620	.7543	.3746	.7992	.2385
13.9649	.1019	.1587	.7525	.3833	.7946	.2218
15.1816	.1052	.1563	.7475	.3965	.7875	.2068
16.6029	.1072	.1542	.7392	.4151	.7776	.1917
18.1835	.1078	.1526	.7289	.4371	.7658	.1773
19.9148	.1072	.1514	.7179	.4602	.7534	.1638
21.8399	.1059	.1504	.7075	.4826	.7414	.1511
23.9276	.1043	.1497	.6989	.5018	.7311	.1393
26.2041	.1028	.1492	.6928	.5167	.7231	.1284
28.4806	.1017	.1487	.6895	.5264	.7179	.1191
31.0421	.1009	.1483	.6883	.5325	.7146	.1101
33.7875	.1004	.1480	.6887	.5356	.7130	.1018
36.7749	.1001	.1476	.6901	.5365	.7125	.0942
40.0665	.1000	.1473	.6920	.5364	.7125	.0869
43.7368	.0999	.1469	.6940	.5359	.7128	.0801
47.8867	.0999	.1467	.6960	.5353	.7131	.0735
52.3614	.0999	.1465	.6977	.5349	.7133	.0676
57.9270	.0998	.1463	.6993	.5348	.7134	.0614
64.6509	.0997	.1461	.7007	.5348	.7134	.0553
73.0210	.0996	.1460	.7021	.5349	.7134	.0492
83.7390	.0995	.1459	.7035	.5349	.7133	.0431
97.6899	.0993	.1458	.7051	.5347	.7135	.0371
113.3524	.0993	.1457	.7065	.5342	.7137	.0321
132.6730	.0993	.1457	.7079	.5336	.7141	.0275
155.2049	.0994	.1456	.7091	.5329	.7144	.0236
201.7717	.0995	.1455	.7107	.5322	.7148	.0183

MACH NO = 25.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0393	.9348	1.3492	-.1340	1.0718	1.0353
.9225	.0417	.8643	1.0994	-.0452	1.0242	.9857
1.0956	.0434	.8052	.9561	.0227	.9879	.9426
1.3972	.0455	.7172	.8115	.1153	.9382	.8759
1.6856	.0465	.6473	.7341	.1839	.9015	.8204
2.0716	.0472	.5703	.6746	.2549	.8634	.7562
2.4207	.0475	.5134	.6441	.3042	.8370	.7062
2.8635	.0477	.4543	.6233	.3515	.8117	.6516
3.2446	.0480	.4127	.6156	.3810	.7958	.6110
3.7079	.0484	.3708	.6144	.4060	.7824	.5679
4.0937	.0491	.3417	.6181	.4194	.7753	.5364
4.4758	.0499	.3172	.6245	.4273	.7710	.5085
4.9268	.0512	.2926	.6344	.4314	.7688	.4791
6.9401	.0603	.2217	.6870	.4135	.7784	.3807
8.6720	.0709	.1900	.7216	.3905	.7907	.3235
10.2140	.0816	.1739	.7427	.3750	.7991	.2854
11.6378	.0911	.1650	.7533	.3695	.8020	.2574
13.1403	.0991	.1593	.7556	.3744	.7994	.2332
14.7468	.1049	.1554	.7509	.3892	.7914	.2119
16.5093	.1078	.1527	.7409	.4120	.7792	.1926
18.4526	.1084	.1508	.7278	.4397	.7644	.1751
20.5841	.1074	.1494	.7144	.4679	.7492	.1592
22.9074	.1056	.1484	.7029	.4929	.7359	.1448
25.4316	.1038	.1478	.6946	.5122	.7255	.1319
28.0761	.1024	.1473	.6901	.5247	.7188	.1206
31.0540	.1015	.1468	.6886	.5320	.7149	.1100
34.3037	.1010	.1464	.6893	.5351	.7133	.1004
37.8555	.1008	.1460	.6912	.5357	.7129	.0917
41.7411	.1007	.1457	.6934	.5353	.7131	.0837
45.9929	.1006	.1454	.6955	.5349	.7133	.0764
50.6457	.1006	.1451	.6972	.5349	.7134	.0697
55.7372	.1004	.1449	.6985	.5351	.7133	.0637
61.3089	.1003	.1448	.6996	.5355	.7130	.0582
67.1867	.1001	.1447	.7006	.5359	.7128	.0533
73.8387	.0999	.1446	.7015	.5362	.7127	.0487
81.1189	.0997	.1446	.7024	.5364	.7126	.0444
89.0869	.0996	.1445	.7033	.5363	.7126	.0406
97.8084	.0995	.1445	.7042	.5362	.7127	.0371
107.3549	.0994	.1445	.7051	.5360	.7128	.0339
117.8050	.0993	.1444	.7060	.5357	.7129	.0309
129.2447	.0992	.1444	.7067	.5354	.7131	.0282
141.3166	.0992	.1444	.7074	.5351	.7132	.0259
154.9825	.0991	.1444	.7081	.5349	.7134	.0236
169.9425	.0991	.1444	.7087	.5346	.7135	.0216
200.4025	.0990	.1444	.7096	.5343	.7137	.0184

MACH NO = 30.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0393	.9342	1.3492	-.1340	1.0718	1.0353
.9221	.0417	.8638	1.0998	-.0454	1.0243	.9858
1.0948	.0434	.8048	.9565	.0224	.9880	.9428
1.3957	.0454	.7170	.8119	.1150	.9384	.8762
1.6834	.0465	.6471	.7343	.1836	.9016	.8207
2.0684	.0472	.5702	.6746	.2546	.8635	.7567
2.4165	.0474	.5133	.6440	.3041	.8371	.7068
2.8578	.0476	.4543	.6229	.3514	.8117	.6523
3.2376	.0478	.4126	.6150	.3812	.7957	.6117
3.6991	.0483	.3707	.6137	.4063	.7823	.5687
4.0832	.0489	.3417	.6173	.4198	.7750	.5372
4.4676	.0497	.3171	.6236	.4279	.7707	.5093
4.9122	.0509	.2925	.6335	.4320	.7685	.4799
5.3117	.0599	.2714	.6861	.4141	.7781	.3818
8.6800	.0708	.1889	.7221	.3900	.7910	.3233
10.1941	.0814	.1731	.7432	.3741	.7995	.2858
11.6878	.0914	.1638	.7545	.3680	.8028	.2565
13.1595	.0994	.1583	.7567	.3729	.8001	.2329
14.7899	.1053	.1544	.7517	.3882	.7920	.2114
16.5210	.1082	.1518	.7415	.4111	.7797	.1925
18.4960	.1087	.1498	.7278	.4398	.7643	.1747
20.5887	.1076	.1484	.7144	.4679	.7492	.1591
22.8647	.1058	.1475	.7028	.4928	.7359	.1451
25.4197	.1039	.1469	.6943	.5127	.7253	.1320
28.1020	.1024	.1464	.6897	.5254	.7184	.1205
31.1221	.1015	.1459	.6883	.5326	.7146	.1098
34.7070	.1010	.1455	.6892	.5353	.7131	.1004
37.9074	.1008	.1451	.6912	.5357	.7129	.0915
41.7136	.1008	.1447	.6935	.5352	.7132	.0837
46.0208	.1007	.1444	.6956	.5348	.7134	.0763
50.5753	.1007	.1442	.6972	.5347	.7135	.0698
55.7294	.1005	.1440	.6986	.5350	.7133	.0637
61.1794	.1004	.1439	.6997	.5354	.7131	.0583
67.1323	.1002	.1438	.7006	.5358	.7128	.0533
73.8694	.1000	.1437	.7015	.5362	.7127	.0486
80.9942	.0998	.1437	.7024	.5364	.7126	.0445
89.0584	.0997	.1436	.7033	.5364	.7126	.0406
97.5877	.0996	.1436	.7042	.5362	.7126	.0372
107.2426	.0995	.1435	.7051	.5360	.7128	.0339
117.4553	.0994	.1435	.7059	.5357	.7129	.0310
129.0167	.0993	.1435	.7067	.5354	.7131	.0283
141.2464	.0993	.1435	.7074	.5351	.7132	.0259
155.0912	.0992	.1435	.7081	.5349	.7134	.0236
169.7362	.0991	.1435	.7087	.5346	.7135	.0216
200.0952	.0991	.1434	.7096	.5343	.7137	.0184

MACH NO = 3.50 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0381	1.0707	1.5198	-.1820	1.1325	1.0642
.8162	.0420	.9893	1.2470	-.1011	1.0736	1.0027
.9552	.0451	.9293	1.1048	-.0478	1.0348	.9543
1.1839	.0493	.8474	.9666	.0175	.9873	.8841
1.3930	.0525	.7866	.8933	.0616	.9552	.8283
1.6651	.0558	.7223	.8353	.1049	.9236	.7655
1.9078	.0583	.6758	.8036	.1343	.9022	.7171
2.2180	.0609	.6274	.7777	.1636	.8809	.6633
2.4984	.0630	.5919	.7633	.1837	.8663	.6213
2.8557	.0655	.5556	.7537	.2022	.8528	.5748
3.1699	.0676	.5300	.7490	.2146	.8438	.5394
3.4973	.0697	.5083	.7470	.2242	.8368	.5068
3.9072	.0722	.4866	.7465	.2334	.8301	.4712
4.4836	.0754	.4637	.7475	.2427	.8233	.4288
5.1786	.0789	.4444	.7494	.2510	.8173	.3868
5.8452	.0817	.4315	.7507	.2577	.8124	.3536
6.5669	.0842	.4216	.7512	.2644	.8075	.3236
7.4593	.0866	.4132	.7505	.2725	.8016	.2928
8.3409	.0882	.4077	.7490	.2802	.7961	.2677
9.3251	.0894	.4034	.7470	.2881	.7902	.2442
10.5832	.0904	.3998	.7443	.2972	.7836	.2197
11.8658	.0908	.3975	.7419	.3051	.7779	.1992
13.3364	.0910	.3956	.7397	.3125	.7725	.1800
15.2618	.0910	.3941	.7379	.3198	.7672	.1599
17.2613	.0909	.3931	.7369	.3252	.7633	.1432
19.5431	.0907	.3923	.7366	.3294	.7602	.1280
22.4295	.0904	.3917	.7370	.3327	.7578	.1128
25.3145	.0903	.3913	.7379	.3347	.7564	.1009
28.5375	.0901	.3910	.7390	.3360	.7554	.0902
32.6177	.0900	.3908	.7404	.3368	.7548	.0795
36.6983	.0899	.3906	.7417	.3372	.7545	.0711
41.2589	.0898	.3905	.7430	.3374	.7544	.0636
47.0343	.0897	.3904	.7443	.3374	.7544	.0561
52.8121	.0897	.3904	.7455	.3373	.7544	.0502
59.2706	.0897	.3903	.7465	.3372	.7545	.0449
67.4509	.0897	.3903	.7476	.3371	.7546	.0396
75.6354	.0897	.3903	.7485	.3370	.7547	.0354
84.7850	.0897	.3903	.7492	.3369	.7548	.0317
96.3745	.0897	.3903	.7500	.3368	.7548	.0280
107.9704	.0897	.3903	.7505	.3367	.7549	.0250
120.9341	.0897	.3903	.7510	.3366	.7549	.0224
137.3550	.0897	.3902	.7515	.3366	.7550	.0197
153.7851	.0897	.3902	.7519	.3366	.7550	.0176
172.1535	.0897	.3902	.7523	.3366	.7550	.0158
200.9993	.0897	.3902	.7527	.3365	.7550	.0135

MACH NO = 5.00 CONE ANGLE = 20.0° ANGLE OF ATTACK = 3.0°

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0381	1.0232	1.5198	-.1820	1.1325	1.0642
.7999	.0410	.9494	1.2672	-.1078	1.0785	1.0387
.9588	.0439	.8792	1.0980	-.0448	1.0326	.9531
1.1715	.0469	.8008	.9650	.0182	.9868	.8876
1.4068	.0496	.7298	.8796	.0699	.9491	.8249
1.6623	.0519	.6669	.8245	.1117	.9187	.7661
1.9343	.0540	.6124	.7895	.1450	.8945	.7121
2.2201	.0560	.5656	.7676	.1710	.8755	.6630
2.5214	.0580	.5253	.7551	.1908	.8611	.6181
2.8335	.0601	.4914	.7493	.2051	.8507	.5775
3.1535	.0624	.4630	.7478	.2152	.8433	.5411
3.4736	.0648	.4394	.7493	.2218	.8385	.5085
3.8111	.0674	.4198	.7522	.2262	.8353	.4791
4.4336	.0722	.3921	.7589	.2310	.8319	.4322
5.0184	.0766	.3738	.7644	.2340	.8297	.3958
5.6264	.0808	.3603	.7680	.2375	.8271	.3639
6.2654	.0844	.3502	.7693	.2424	.8235	.3355
7.0178	.0878	.3419	.7686	.2494	.8184	.3073
7.7592	.0903	.3364	.7663	.2573	.8127	.2837
8.5753	.0922	.3322	.7626	.2663	.8061	.2617
9.4860	.0934	.3290	.7579	.2765	.7987	.2408
10.5164	.0941	.3265	.7526	.2874	.7908	.2209
11.8252	.0943	.3245	.7465	.2995	.7820	.1998
13.2159	.0940	.3230	.7416	.3099	.7744	.1815
14.8459	.0936	.3218	.7377	.3191	.7678	.1638
16.7699	.0930	.3208	.7352	.3264	.7624	.1470
18.0511	.0925	.3200	.7342	.3316	.7586	.1310
22.0308	.0921	.3193	.7347	.3352	.7560	.1147
25.1179	.0918	.3188	.7360	.3369	.7548	.1016
28.5964	.0917	.3184	.7377	.3376	.7542	.0900
32.5172	.0915	.3181	.7395	.3378	.7541	.0798
37.4092	.0915	.3179	.7415	.3378	.7541	.0699
42.4536	.0914	.3178	.7431	.3376	.7542	.0619
48.1428	.0914	.3177	.7445	.3375	.7543	.0549
54.5598	.0914	.3176	.7458	.3373	.7544	.0486
61.7984	.0913	.3176	.7469	.3372	.7546	.0431
70.8361	.0913	.3175	.7480	.3370	.7547	.0378
80.1538	.0913	.3175	.7489	.3369	.7547	.0335
90.6781	.0913	.3175	.7496	.3368	.7548	.0297
102.5445	.0913	.3175	.7503	.3368	.7549	.0263
115.9315	.0913	.3175	.7508	.3367	.7549	.0233
132.6472	.0913	.3175	.7514	.3366	.7549	.0204
149.8922	.0913	.3175	.7518	.3366	.7550	.0181
169.3474	.0913	.3175	.7522	.3366	.7550	.0160
200.8437	.0913	.3175	.7526	.3366	.7550	.0135

MACH NO = 10.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0380	.9861	1.5198	-.1820	1.1375	1.0642
.8140	.0407	.9355	1.2472	-.1010	1.0735	1.0035
.9929	.0432	.8276	1.0674	-.0315	1.0229	.9420
1.2288	.0456	.7427	.9325	.0361	.9737	.8715
1.4903	.0475	.6664	.8481	.0911	.9337	.8147
1.7717	.0491	.6001	.7960	.1344	.9022	.7435
2.0232	.0504	.5514	.7636	.1630	.8814	.6961
2.3250	.0518	.5031	.7498	.1881	.8631	.6466
2.6319	.0536	.4630	.7418	.2049	.8518	.6031
2.9391	.0555	.4301	.7407	.2153	.8433	.5650
3.2439	.0578	.4032	.7441	.2207	.8394	.5317
3.5021	.0600	.3840	.7481	.2225	.8381	.5064
3.7992	.0628	.3654	.7561	.2224	.8381	.4801
4.4232	.0692	.3359	.7710	.2192	.8414	.4331
5.0245	.0755	.3159	.7818	.2168	.8422	.3954
5.5809	.0810	.3030	.7881	.2165	.8424	.3661
6.1809	.0864	.2932	.7914	.2181	.8412	.3395
6.8004	.0912	.2863	.7935	.2222	.8382	.3149
7.4557	.0951	.2814	.7900	.2293	.8331	.2929
8.1152	.0978	.2780	.7850	.2388	.8262	.2737
8.8922	.0994	.2753	.7769	.2520	.8166	.2540
9.7633	.0999	.2733	.7668	.2677	.8052	.2351
10.7472	.0994	.2718	.7556	.2847	.7927	.2168
11.7773	.0983	.2706	.7457	.3003	.7814	.2005
13.0123	.0969	.2696	.7370	.3148	.7709	.1839
14.3868	.0957	.2686	.7314	.3255	.7630	.1684
15.7983	.0948	.2678	.7289	.3320	.7583	.1550
17.4734	.0942	.2670	.7285	.3359	.7555	.1416
19.3672	.0939	.2663	.7298	.3377	.7542	.1290
21.5470	.0937	.2656	.7319	.3382	.7538	.1171
23.9219	.0936	.2651	.7341	.3382	.7538	.1063
26.9585	.0936	.2646	.7364	.3380	.7540	.0951
30.6965	.0935	.2642	.7387	.3378	.7541	.0842
35.4140	.0935	.2639	.7409	.3376	.7542	.0736
40.9558	.0934	.2637	.7428	.3375	.7543	.0641
47.9011	.0934	.2636	.7446	.3372	.7545	.0551
55.9694	.0934	.2635	.7462	.3370	.7547	.0475
64.6708	.0934	.2634	.7475	.3369	.7548	.0413
75.4511	.0934	.2634	.7486	.3367	.7549	.0355
87.9779	.0934	.2633	.7496	.3366	.7549	.0306
102.5315	.0934	.2633	.7504	.3366	.7550	.0263
118.2353	.0934	.2633	.7510	.3365	.7550	.0229
137.6908	.0934	.2633	.7516	.3365	.7550	.0197
160.2974	.0934	.2633	.7521	.3365	.7550	.0169
200.2401	.0934	.2633	.7527	.3365	.7550	.0136

MACH NO = 15.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0380	.9790	1.5198	-.1820	1.1325	1.0642
.8384	.0410	.8870	1.2162	-.0902	1.0656	.9947
1.0171	.0432	.8108	1.0487	-.0231	1.0168	.9342
1.2516	.0454	.7278	.9214	.0426	.9690	.8652
1.5103	.0470	.6533	.8410	.0961	.9331	.8001
1.7868	.0484	.5886	.7910	.1382	.8994	.7404
2.0739	.0497	.5338	.7613	.1701	.8762	.6873
2.3674	.0510	.4880	.7449	.1933	.8593	.6403
2.6626	.0526	.4500	.7384	.2086	.8481	.5990
2.9557	.0544	.4187	.7384	.2177	.8415	.5631
3.2444	.0566	.3930	.7426	.2219	.8384	.5316
3.5273	.0591	.3719	.7493	.2229	.8378	.5040
3.8427	.0622	.3523	.7581	.2215	.8388	.4765
4.4862	.0692	.3223	.7751	.2162	.8426	.4286
5.0646	.0756	.3036	.7862	.2128	.8451	.3931
5.6616	.0820	.2902	.7940	.2113	.8462	.3622
6.2572	.0880	.2810	.7982	.2120	.8457	.3358
6.8696	.0932	.2747	.7989	.2158	.8429	.3124
7.5165	.0974	.2701	.7958	.2233	.8374	.2910
8.2170	.1002	.2669	.7892	.2346	.8292	.2709
8.9899	.1016	.2645	.7794	.2496	.8183	.2517
9.8495	.1016	.2628	.7673	.2674	.8053	.2334
10.8036	.1005	.2615	.7544	.2864	.7915	.2159
11.8591	.0989	.2605	.7427	.3042	.7786	.1993
13.0155	.0972	.2595	.7338	.3186	.7681	.1839
14.2752	.0958	.2586	.7285	.3287	.7607	.1696
15.6427	.0949	.2578	.7264	.3347	.7564	.1564
17.1429	.0945	.2570	.7268	.3374	.7544	.1441
18.8106	.0943	.2563	.7287	.3382	.7538	.1325
20.6854	.0942	.2556	.7310	.3381	.7539	.1215
22.6849	.0942	.2551	.7332	.3378	.7541	.1116
25.1310	.0941	.2546	.7354	.3377	.7542	.1016
28.0268	.0941	.2542	.7373	.3377	.7542	.0917
31.5680	.0940	.2539	.7392	.3377	.7542	.0820
36.0566	.0940	.2536	.7412	.3375	.7543	.0723
41.9215	.0940	.2534	.7433	.3372	.7545	.0627
49.5300	.0940	.2532	.7454	.3367	.7549	.0534
58.5417	.0941	.2530	.7472	.3363	.7552	.0454
69.1301	.0942	.2530	.7487	.3359	.7555	.0387
81.5724	.0943	.2529	.7498	.3358	.7555	.0329
96.1946	.0944	.2528	.7506	.3359	.7555	.0280
113.3790	.0944	.2528	.7513	.3360	.7554	.0238
133.5747	.0944	.2528	.7518	.3361	.7554	.0203
157.3093	.0945	.2528	.7523	.3362	.7553	.0173
201.8975	.0945	.2528	.7529	.3363	.7552	.0135

MACH NO = 20.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6582	.0379	.9763	1.5138	-.1820	1.1325	1.0642
.8369	.0409	.8850	1.2179	-.0907	1.0661	.9952
1.0142	.0431	.8092	1.0505	-.0239	1.0174	.9351
1.2469	.0452	.7265	.9228	.0417	.9696	.8665
1.5415	.0470	.6423	.8330	.1020	.9257	.7929
1.8173	.0483	.5792	.7859	.1428	.8960	.7344
2.1023	.0495	.5258	.7581	.1736	.8737	.6824
2.3896	.0508	.4816	.7434	.1953	.8578	.6370
2.7144	.0525	.4407	.7374	.2111	.8463	.5924
2.9923	.0544	.4119	.7384	.2187	.8408	.5589
3.2628	.0564	.3881	.7430	.2221	.8383	.5297
3.5250	.0588	.3686	.7495	.2226	.8380	.5042
3.8143	.0617	.3504	.7579	.2210	.8391	.4788
4.9391	.0742	.3034	.7857	.2119	.8458	.4003
5.9803	.0857	.2811	.7989	.2090	.8478	.3476
6.9692	.0947	.2702	.8010	.2142	.8440	.3089
8.0402	.1006	.2641	.7930	.2294	.8330	.2757
9.1823	.1026	.2606	.7777	.2524	.8162	.2474
10.4915	.1016	.2585	.7583	.2806	.7957	.2213
11.9019	.0992	.2570	.7414	.3058	.7774	.1987
13.5017	.0969	.2558	.7304	.3241	.7640	.1781
15.2097	.0956	.2547	.7267	.3333	.7574	.1604
17.1441	.0952	.2537	.7278	.3362	.7552	.1441
19.2171	.0952	.2529	.7309	.3361	.7554	.1300
21.5774	.0954	.2522	.7340	.3355	.7558	.1169
24.1169	.0954	.2517	.7362	.3356	.7557	.1055
27.0141	.0953	.2514	.7377	.3363	.7552	.0949
30.1337	.0951	.2511	.7388	.3373	.7545	.0857
33.6976	.0949	.2510	.7397	.3382	.7538	.0771
37.6460	.0947	.2509	.7407	.3389	.7533	.0694
41.9024	.0945	.2508	.7417	.3392	.7531	.0627
46.7604	.0943	.2507	.7429	.3393	.7530	.0564
51.9923	.0942	.2507	.7440	.3392	.7531	.0510
57.9639	.0941	.2506	.7451	.3390	.7532	.0459
64.7960	.0940	.2506	.7461	.3387	.7534	.0414
71.7385	.0939	.2506	.7470	.3385	.7536	.0373
79.6479	.0939	.2505	.7478	.3383	.7538	.0337
88.6773	.0938	.2505	.7486	.3381	.7539	.0303
98.4040	.0938	.2505	.7492	.3379	.7540	.0274
109.5078	.0937	.2505	.7498	.3377	.7542	.0247
121.4690	.0937	.2505	.7504	.3376	.7543	.0223
135.1236	.0937	.2505	.7509	.3374	.7544	.0200
149.8326	.0936	.2505	.7513	.3373	.7545	.0181
166.6242	.0936	.2505	.7517	.3372	.7545	.0163
207.4499	.0936	.2505	.7523	.3370	.7547	.0136

MACH NO = 25.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0379	.9153	1.5198	-.1820	1.1325	1.0642
.8367	.0429	.8842	1.2186	-.0910	1.0662	.9954
1.0129	.0430	.8096	1.0513	-.0243	1.0177	.9356
1.2447	.0451	.7261	.9234	.0413	.9699	.8671
1.5391	.0469	.6420	.8334	.1017	.9260	.7936
1.8126	.0481	.5788	.7860	.1426	.8962	.7353
2.0952	.0493	.5255	.7579	.1734	.8738	.6835
2.3819	.0506	.4813	.7430	.1953	.8578	.6381
2.7047	.0523	.4403	.7369	.2112	.8463	.5936
2.9808	.0541	.4114	.7378	.2189	.8417	.5602
3.2492	.0561	.3876	.7424	.2223	.8382	.5311
3.5092	.0584	.3680	.7490	.2228	.8379	.5057
3.7957	.0613	.3498	.7575	.2211	.8390	.4804
4.0371	.0741	.3016	.7862	.2113	.8462	.4005
5.0571	.0856	.2796	.7997	.2079	.8486	.3486
6.0798	.0951	.2683	.8022	.2130	.8449	.3086
8.0259	.1010	.2624	.7942	.2280	.8340	.2761
9.1748	.1030	.2590	.7784	.2517	.8168	.2476
10.4927	.1018	.2568	.7581	.2808	.7956	.2213
11.9142	.0992	.2554	.7406	.3068	.7767	.1985
13.4780	.0969	.2542	.7297	.3249	.7635	.1784
15.2455	.0956	.2530	.7261	.3341	.7568	.1600
17.1361	.0952	.2525	.7276	.3365	.7591	.1441
19.2200	.0953	.2512	.7309	.3360	.7554	.1299
21.5920	.0955	.2505	.7342	.3353	.7559	.1168
24.1476	.0955	.2500	.7364	.3354	.7559	.1054
26.9774	.0954	.2497	.7378	.3362	.7553	.0951
30.2052	.0952	.2495	.7388	.3373	.7545	.0855
33.6856	.0950	.2493	.7397	.3383	.7538	.0772
37.5401	.0949	.2492	.7406	.3389	.7533	.0696
41.9378	.0945	.2491	.7417	.3392	.7530	.0626
46.6803	.0944	.2490	.7428	.3393	.7530	.0565
51.9332	.0943	.2490	.7440	.3392	.7531	.0510
57.9270	.0942	.2489	.7451	.3390	.7533	.0459
64.3915	.0941	.2489	.7461	.3387	.7534	.0414
71.5531	.0940	.2489	.7470	.3385	.7536	.0374
79.7261	.0939	.2488	.7478	.3383	.7538	.0336
88.5421	.0939	.2488	.7486	.3381	.7539	.0304
98.3093	.0939	.2488	.7492	.3379	.7540	.0274
109.4560	.0938	.2488	.7498	.3377	.7542	.0247
121.4793	.0938	.2488	.7504	.3376	.7543	.0223
134.7995	.0939	.2488	.7509	.3374	.7544	.0201
150.0010	.0937	.2488	.7513	.3373	.7545	.0181
166.7997	.0937	.2488	.7517	.3372	.7545	.0163
200.4994	.0937	.2488	.7523	.3370	.7547	.0136

MACH NO = 30.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 3.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0373	.9746	1.5198	-.1820	1.1325	1.0642
.8359	.0408	.8837	1.2131	-.0912	1.0664	.9956
1.0121	.0431	.8282	1.0517	-.0245	1.0178	.9358
1.2434	.0460	.7258	.9238	.0411	.9701	.8675
1.5362	.0468	.6417	.8336	.1015	.9261	.7941
1.8100	.0480	.5786	.7861	.1425	.8963	.7358
2.0928	.0492	.5253	.7578	.1733	.8738	.6840
2.3776	.0504	.4811	.7428	.1953	.8578	.6387
2.6993	.0521	.4401	.7356	.2112	.8462	.5943
2.9744	.0539	.4111	.7375	.2190	.8406	.5609
3.2416	.0559	.3873	.7420	.2224	.8381	.5319
3.5004	.0582	.3677	.7487	.2228	.8378	.5065
3.7854	.0611	.3494	.7572	.2211	.8390	.4812
4.9491	.0742	.3003	.7868	.2109	.8465	.3998
5.9566	.0857	.2796	.8003	.2073	.8491	.3486
6.9657	.0952	.2674	.8029	.2122	.8456	.3091
8.0274	.1012	.2614	.7948	.2274	.8344	.2761
9.1962	.1032	.2590	.7783	.2519	.8167	.2471
10.4622	.1020	.2560	.7585	.2803	.7960	.2218
11.9040	.0993	.2545	.7403	.3071	.7765	.1987
13.4937	.0969	.2533	.7292	.3256	.7630	.1782
15.2389	.0956	.2522	.7258	.3345	.7565	.1601
17.1590	.0952	.2511	.7275	.3366	.7550	.1440
19.2167	.0953	.2503	.7310	.3359	.7555	.1300
21.5559	.0955	.2496	.7343	.3351	.7560	.1170
24.1487	.0956	.2491	.7365	.3352	.7560	.1054
27.0230	.0955	.2488	.7379	.3361	.7563	.0949
30.2128	.0953	.2485	.7389	.3373	.7545	.0855
33.6465	.0951	.2484	.7397	.3383	.7538	.0772
37.5594	.0948	.2483	.7406	.3389	.7533	.0696
41.8970	.0946	.2482	.7417	.3393	.7530	.0627
46.7110	.0944	.2481	.7428	.3393	.7530	.0565
52.0523	.0943	.2481	.7440	.3392	.7531	.0509
57.8838	.0942	.2480	.7450	.3390	.7533	.0460
64.3583	.0941	.2480	.7461	.3387	.7534	.0415
71.6308	.0941	.2479	.7470	.3385	.7536	.0374
79.7004	.0940	.2479	.7478	.3383	.7538	.0337
88.3931	.0939	.2479	.7486	.3381	.7539	.0304
98.3001	.0939	.2479	.7492	.3379	.7540	.0274
109.2930	.0939	.2479	.7498	.3377	.7542	.0247
121.4907	.0938	.2479	.7504	.3376	.7543	.0223
135.0252	.0938	.2479	.7509	.3374	.7544	.0201
149.6045	.0938	.2479	.7513	.3373	.7545	.0181
166.2206	.0938	.2479	.7517	.3372	.7545	.0163
200.3847	.0937	.2479	.7523	.3370	.7547	.0136

MACH NO = 3.50 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8411	.0701	.9934	1.1889	-.0805	1.0141	1.0129
1.0879	.0783	.9544	.9275	.0390	.9932	.9886
1.3455	.0862	.9200	.7788	.1452	.9748	.9671
1.8554	.0994	.8579	.6353	.3137	.9451	.9271
2.4125	.1102	.7979	.5644	.4661	.9184	.8870
3.2608	.1211	.7196	.5136	.6600	.8845	.8322
4.1304	.1281	.6524	.4898	.8247	.8557	.7827
5.3827	.1345	.5738	.4777	1.0133	.8227	.7208
6.6097	.1385	.5126	.4770	1.1566	.7976	.6691
8.3139	.1421	.4466	.4831	1.3073	.7713	.6084
9.9371	.1443	.3985	.4922	1.4129	.7528	.5600
11.7547	.1462	.3567	.5035	1.5006	.7374	.5142
14.2066	.1480	.3144	.5183	1.5847	.7227	.4631
16.0141	.1491	.2905	.5284	1.6296	.7149	.4315
17.9672	.1501	.2698	.5383	1.6669	.7083	.4019
20.0725	.1510	.2518	.5480	1.6976	.7030	.3742
22.3372	.1519	.2363	.5572	1.7227	.6986	.3484
24.7691	.1528	.2229	.5660	1.7431	.6950	.3243
27.3763	.1536	.2114	.5744	1.7595	.6921	.3020
30.1676	.1545	.2015	.5821	1.7726	.6898	.2812
33.1525	.1554	.1929	.5894	1.7832	.5880	.2620
36.3410	.1562	.1856	.5962	1.7916	.6865	.2442
39.7438	.1571	.1792	.6024	1.7986	.6853	.2276
43.3722	.1579	.1738	.6081	1.8042	.6843	.2123
47.2384	.1587	.1691	.6133	1.8090	.6835	.1981
51.3555	.1595	.1651	.6180	1.8131	.6827	.1849
55.7373	.1603	.1616	.6224	1.8167	.6821	.1726
60.3987	.1610	.1586	.6263	1.8200	.6815	.1613
65.3558	.1617	.1559	.6299	1.8230	.6810	.1507
70.6256	.1624	.1537	.6332	1.8257	.6805	.1409
76.2268	.1630	.1517	.6361	1.8284	.6801	.1318
83.7253	.1637	.1497	.6394	1.8315	.6795	.1213
90.1485	.1642	.1483	.6418	1.8339	.6791	.1136
96.9750	.1647	.1470	.6440	1.8363	.6787	.1064
104.2312	.1652	.1460	.6460	1.8386	.6783	.0996
111.9454	.1656	.1450	.6478	1.8408	.6779	.0934
120.1485	.1660	.1442	.6494	1.8430	.6775	.0875
128.8736	.1663	.1435	.6509	1.8452	.6771	.0820
138.1561	.1667	.1429	.6523	1.8472	.6768	.0769
148.0342	.1670	.1424	.6535	1.8492	.6764	.0721
158.5482	.1672	.1419	.6547	1.8511	.6761	.0676
169.7416	.1675	.1415	.6558	1.8530	.6758	.0634
181.6601	.1677	.1411	.6568	1.8547	.6755	.0595
194.3525	.1679	.1408	.6577	1.8563	.6752	.0558
201.0049	.1680	.1407	.6581	1.8571	.6750	.0540

MACH NO = 5.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8533	.0693	.9436	1.1719	-.0742	1.0130	1.0109
1.0579	.0750	.9123	.9507	.0258	.9955	.9912
1.3361	.0817	.8754	.7791	.1429	.9750	.9678
1.8081	.0909	.8181	.6334	.3085	.9460	.9306
2.4323	.0990	.7509	.5448	.4902	.9142	.8856
3.2403	.1050	.6761	.4908	.6875	.8797	.8335
4.2533	.1089	.5980	.4611	.8895	.8444	.7761
5.4614	.1114	.5227	.4501	1.0771	.8115	.7173
6.8698	.1132	.4533	.4525	1.2393	.7831	.6590
8.4872	.1150	.3915	.4645	1.3698	.7603	.6028
10.3028	.1172	.3383	.4824	1.4669	.7433	.5501
11.8918	.1193	.3019	.4988	1.5231	.7335	.5111
14.0460	.1224	.2634	.5198	1.5720	.7249	.4662
16.3847	.1260	.2317	.5400	1.6036	.7194	.4256
18.9070	.1298	.2058	.5584	1.6241	.7158	.3890
21.0571	.1330	.1885	.5714	1.6358	.7138	.3625
23.9110	.1369	.1705	.5855	1.6475	.7117	.3324
26.3286	.1399	.1585	.5950	1.6557	.7103	.3106
29.5208	.1436	.1459	.6051	1.6658	.7085	.2858
32.9053	.1469	.1356	.6134	1.6761	.7067	.2635
35.7541	.1494	.1286	.6189	1.6845	.7052	.2473
39.4953	.1522	.1213	.6247	1.6953	.7034	.2287
42.6361	.1543	.1163	.6286	1.7040	.7018	.2152
46.7528	.1566	.1110	.6327	1.7148	.6999	.1997
51.0901	.1587	.1066	.6362	1.7255	.6981	.1857
54.7276	.1602	.1036	.6386	1.7338	.6966	.1753
59.4990	.1620	.1003	.6411	1.7439	.6949	.1634
63.5099	.1632	.0980	.6429	1.7517	.6935	.1545
68.7890	.1646	.0956	.6449	1.7611	.6918	.1442
74.3919	.1658	.0935	.6467	1.7701	.6903	.1347
79.1230	.1667	.0920	.6479	1.7771	.6890	.1276
85.3962	.1677	.0904	.6493	1.7854	.6876	.1192
92.0776	.1686	.0890	.6505	1.7933	.6862	.1115
97.7420	.1692	.0880	.6514	1.7992	.6852	.1056
105.2486	.1699	.0869	.6525	1.8061	.6840	.0988
111.6148	.1704	.0862	.6533	1.8113	.6831	.0936
120.0526	.1710	.0854	.6543	1.8173	.6820	.0876
129.0574	.1715	.0846	.6552	1.8228	.6811	.0819
136.6953	.1719	.0841	.6559	1.8268	.6804	.0777
146.8193	.1723	.0836	.6567	1.8314	.6795	.0727
155.4069	.1726	.0832	.6573	1.8348	.6789	.0689
166.7898	.1730	.0828	.6581	1.8387	.6783	.0645
178.9382	.1733	.0824	.6588	1.8422	.6776	.0603
189.2422	.1735	.0822	.6593	1.8449	.6772	.0572
200.0963	.1738	.0819	.6598	1.8473	.6768	.0543

MACH NO = 10.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8620	.0684	.9052	1.1681	-.1697	1.0122	1.0097
1.1279	.0744	.8669	.8960	.0578	.9899	.9852
1.4400	.0800	.8270	.7314	.1855	.9675	.9594
2.0603	.0873	.7553	.5766	.3977	.9304	.9119
2.7452	.0912	.6864	.4987	.5949	.8959	.8647
3.5796	.0925	.6145	.4499	.8007	.8599	.8133
4.7790	.0917	.5296	.4170	1.0440	.8173	.7494
5.9345	.0898	.4639	.4051	1.2296	.7848	.6966
7.2182	.0877	.4047	.4043	1.3889	.7570	.6461
8.9072	.0855	.3434	.4142	1.5387	.7308	.5898
10.4226	.0842	.3002	.4286	1.6288	.7150	.5470
12.0249	.0835	.2635	.4456	1.6903	.7042	.5080
14.0532	.0836	.2267	.4712	1.7317	.6970	.4660
16.1929	.0845	.1966	.4972	1.7448	.6947	.4286
18.8336	.0869	.1686	.5278	1.7341	.6966	.3900
21.2276	.0899	.1481	.5531	1.7102	.7008	.3606
23.7481	.0937	.1315	.5766	1.6790	.7062	.3340
26.8492	.0991	.1157	.6010	1.6403	.7130	.3062
29.6386	.1043	.1046	.6189	1.6095	.7184	.2850
32.5388	.1098	.0953	.6337	1.5838	.7229	.2657
36.0538	.1163	.0864	.6473	1.5617	.7267	.2457
39.1916	.1219	.0801	.6564	1.5496	.7289	.2301
42.4910	.1274	.0747	.6634	1.5431	.7300	.2158
46.5840	.1336	.0693	.6694	1.5424	.7301	.2003
50.2301	.1384	.0655	.6728	1.5471	.7293	.1883
54.6897	.1435	.0617	.6751	1.5583	.7273	.1754
58.7210	.1474	.0589	.6760	1.5716	.7250	.1652
62.9721	.1508	.0564	.6761	1.5873	.7223	.1556
68.2410	.1543	.0539	.6756	1.6072	.7188	.1452
73.0471	.1570	.0520	.6749	1.6248	.7157	.1369
78.1414	.1594	.0503	.6741	1.6424	.7126	.1290
84.4711	.1619	.0487	.6730	1.6626	.7091	.1204
90.2431	.1637	.0474	.6720	1.6795	.7061	.1135
96.3498	.1653	.0463	.6710	1.6959	.7033	.1070
103.9218	.1669	.0452	.6699	1.7140	.7001	.0999
110.8203	.1681	.0443	.6689	1.7287	.6975	.0942
119.3762	.1693	.0434	.6679	1.7449	.6947	.0880
127.1705	.1701	.0428	.6671	1.7580	.6924	.0830
135.4152	.1708	.0422	.6663	1.7701	.6903	.0783
145.6373	.1714	.0416	.6655	1.7832	.6880	.0732
154.9482	.1719	.0411	.6650	1.7934	.6862	.0691
164.7980	.1723	.0407	.6645	1.8028	.6846	.0652
177.0146	.1727	.0403	.6641	1.8125	.6829	.0610
188.1474	.1729	.0400	.6638	1.8199	.6816	.0575
201.9597	.1732	.0397	.6636	1.8276	.6802	.0538

MACH NO = 15.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	ΔVCP/LV	RN/RB
.8636	.0683	.8979	1.1579	-.0688	1.0120	1.0094
1.1240	.0739	.8606	.9986	.0561	.9902	.9855
1.5189	.0804	.8106	.7017	.2159	.9622	.9531
2.0430	.0858	.7506	.5767	.3948	.9309	.9132
2.8611	.0894	.6693	.4851	.6314	.8895	.8572
3.6996	.0897	.5988	.4385	.8376	.8534	.8064
4.8897	.0878	.5166	.4056	1.0812	.8108	.7440
6.0210	.0851	.4534	.3924	1.2676	.7782	.6930
7.5209	.0817	.3864	.3896	1.4579	.7449	.6352
8.8676	.0792	.3386	.3955	1.5841	.7228	.5910
10.5745	.0767	.2903	.4090	1.6969	.7031	.5430
12.0557	.0752	.2567	.4239	1.7617	.6917	.5073
13.8869	.0741	.2231	.4445	1.8098	.6833	.4692
16.3947	.0737	.1877	.4741	1.8341	.6791	.4254
18.9530	.0745	.1603	.5040	1.8255	.6806	.3884
21.5403	.0764	.1390	.5331	1.7955	.6858	.3570
24.1394	.0792	.1223	.5603	1.7526	.6933	.3302
26.7361	.0829	.1090	.5851	1.7038	.7019	.3072
29.3196	.0873	.0984	.6070	1.6547	.7105	.2872
31.8847	.0922	.0898	.6260	1.6091	.7184	.2698
34.4374	.0974	.0827	.6419	1.5694	.7254	.2545
36.6812	.1021	.0775	.6536	1.5402	.7305	.2424
39.3027	.1077	.0724	.6647	1.5131	.7352	.2296
42.0529	.1135	.0679	.6737	1.4931	.7387	.2176
45.0196	.1193	.0638	.6806	1.4808	.7409	.2060
48.2590	.1250	.0601	.6854	1.4773	.7415	.1946
51.8304	.1305	.0567	.6881	1.4827	.7406	.1835
55.7936	.1355	.0535	.6891	1.4960	.7382	.1725
60.2756	.1403	.0506	.6888	1.5155	.7348	.1616
65.3084	.1446	.0479	.6875	1.5390	.7307	.1508
70.7268	.1485	.0455	.6858	1.5639	.7264	.1408
76.5260	.1519	.0435	.6838	1.5893	.7219	.1314
82.7116	.1548	.0417	.6817	1.6146	.7175	.1227
89.3048	.1573	.0402	.6796	1.6390	.7132	.1146
96.3350	.1594	.0388	.6775	1.6625	.7091	.1070
103.8338	.1612	.0376	.6755	1.6847	.7052	.1000
111.8339	.1626	.0366	.6736	1.7055	.7016	.0935
120.3714	.1639	.0357	.6719	1.7248	.6982	.0874
128.3129	.1648	.0350	.6706	1.7404	.6955	.0824
137.9623	.1656	.0343	.6693	1.7567	.6926	.0770
148.2617	.1664	.0337	.6682	1.7713	.6901	.0720
159.2549	.1670	.0332	.6673	1.7842	.6878	.0673
170.9909	.1675	.0327	.6665	1.7957	.6858	.0630
183.5237	.1679	.0323	.6660	1.8058	.6840	.0589
200.4011	.1684	.0318	.6655	1.8167	.6821	.0542

MACH NO = 20.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8642	.0682	.8951	1.1572	-.0685	1.0120	1.0094
1.1225	.0737	.8582	.8996	.0555	.9903	.9857
1.5151	.0800	.8086	.7026	.2148	.9624	.9534
2.0359	.0852	.7489	.5770	.3935	.9312	.9137
2.8459	.0886	.6683	.4846	.6294	.8899	.8581
3.6746	.0887	.5984	.4371	.8355	.8538	.8079
4.8480	.0866	.5169	.4029	1.0798	.8111	.7460
5.9603	.0837	.4542	.3892	1.2681	.7781	.6955
7.4296	.0800	.3879	.3835	1.4622	.7441	.6384
8.7431	.0771	.3405	.3875	1.5927	.7213	.5948
10.3990	.0743	.2926	.3998	1.7117	.7005	.5476
11.8274	.0724	.2593	.4119	1.7824	.6881	.5125
13.5806	.0708	.2261	.4304	1.8379	.6784	.4752
16.2525	.0697	.1872	.4607	1.8744	.6720	.4277
18.9309	.0698	.1582	.4915	1.8711	.6726	.3887
21.5759	.0710	.1364	.5212	1.8423	.6776	.3566
24.4412	.0735	.1181	.5519	1.7927	.6863	.3274
26.9357	.0766	.1056	.5771	1.7402	.6955	.3055
29.3443	.0804	.0957	.5996	1.6862	.7050	.2871
31.6719	.0847	.0878	.6194	1.6345	.7140	.2712
33.9350	.0893	.0814	.6365	1.5876	.7222	.2574
36.1651	.0943	.0760	.6511	1.5464	.7294	.2451
38.6615	.1000	.0710	.6646	1.5080	.7361	.2326
40.9889	.1054	.0670	.6747	1.4806	.7409	.2221
43.4614	.1109	.0633	.6828	1.4608	.7444	.2119
46.1511	.1164	.0599	.6888	1.4498	.7463	.2019
49.0591	.1216	.0568	.6925	1.4480	.7466	.1920
52.2384	.1266	.0538	.6944	1.4548	.7454	.1823
55.7997	.1313	.0510	.6947	1.4689	.7430	.1725
60.3745	.1363	.0480	.6938	1.4913	.7391	.1613
65.2331	.1407	.0454	.6921	1.5162	.7347	.1510
70.8161	.1448	.0430	.6898	1.5445	.7298	.1406
76.9096	.1484	.0409	.6871	1.5737	.7246	.1308
83.4057	.1514	.0390	.6844	1.6023	.7196	.1218
90.3389	.1540	.0374	.6817	1.6298	.7148	.1134
98.6075	.1564	.0358	.6789	1.6588	.7097	.1048
106.6053	.1582	.0346	.6766	1.6831	.7055	.0976
115.1749	.1596	.0336	.6744	1.7055	.7016	.0910
124.3570	.1608	.0327	.6726	1.7258	.6980	.0848
134.1908	.1619	.0319	.6710	1.7442	.6948	.0790
144.7175	.1627	.0312	.6697	1.7605	.6919	.0737
155.9849	.1635	.0306	.6686	1.7750	.6894	.0687
169.4412	.1642	.0300	.6676	1.7890	.6870	.0635
182.4661	.1647	.0295	.6670	1.8000	.6850	.0592
201.3112	.1653	.0290	.6663	1.8126	.6828	.0540

MACH NO = 25.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8645	.0682	.8941	1.1568	-.0684	1.0120	1.0093
1.1218	.0735	.8573	.9001	.0552	.9903	.9857
1.5133	.0798	.8078	.7030	.2143	.9625	.9535
2.0326	.0853	.7483	.5771	.3928	.9313	.9139
2.8387	.0883	.6680	.4843	.6284	.8900	.8586
3.8441	.0881	.5946	.4292	.8758	.8468	.7983
4.8284	.0860	.5172	.4016	1.0791	.8112	.7470
6.1651	.0824	.4431	.3844	1.3036	.7719	.6870
7.6407	.0786	.3787	.3807	1.4930	.7388	.6310
8.9515	.0757	.3329	.3850	1.6198	.7166	.5884
10.5950	.0728	.2867	.3962	1.7354	.6963	.5425
12.0052	.0708	.2547	.4089	1.8042	.6843	.5085
13.7269	.0691	.2227	.4266	1.8587	.6748	.4723
15.6200	.0677	.1917	.4508	1.8973	.6680	.4219
19.4814	.0677	.1522	.4911	1.8913	.6691	.3816
22.2572	.0689	.1306	.5219	1.8580	.6749	.3492
24.9132	.0710	.1144	.5504	1.8090	.6835	.3230
27.4351	.0741	.1022	.5763	1.7524	.6934	.3015
29.8248	.0777	.0928	.5994	1.6944	.7035	.2836
32.0974	.0820	.0853	.6197	1.6388	.7133	.2685
34.2809	.0865	.0793	.6373	1.5880	.7221	.2554
36.4162	.0914	.0743	.6524	1.5430	.7300	.2438
38.5550	.0966	.0700	.6652	1.5046	.7367	.2331
40.7575	.1019	.0662	.6760	1.4733	.7422	.2231
43.0920	.1074	.0627	.6846	1.4501	.7463	.2134
45.6278	.1128	.0594	.6910	1.4362	.7487	.2037
48.3464	.1181	.0564	.6951	1.4323	.7494	.1943
51.3122	.1229	.0535	.6971	1.4375	.7485	.1850
54.6407	.1276	.0507	.6975	1.4506	.7462	.1755
58.4798	.1321	.0481	.6968	1.4698	.7428	.1658
62.9730	.1365	.0454	.6951	1.4938	.7386	.1556
67.9731	.1407	.0430	.6930	1.5204	.7340	.1457
73.6231	.1445	.0408	.6903	1.5495	.7289	.1359
79.5021	.1477	.0389	.6875	1.5778	.7239	.1270
86.6034	.1507	.0370	.6844	1.6091	.7184	.1177
94.2277	.1531	.0354	.6813	1.6390	.7132	.1092
102.4336	.1552	.0340	.6785	1.6669	.7083	.1012
111.2758	.1569	.0328	.6760	1.6923	.7039	.0939
120.8019	.1583	.0317	.6738	1.7153	.6999	.0871
131.0525	.1595	.0308	.6720	1.7358	.6963	.0808
142.0715	.1605	.0300	.6705	1.7540	.6931	.0749
153.9151	.1614	.0294	.6692	1.7699	.6903	.0695
166.6539	.1621	.0288	.6683	1.7838	.6879	.0645
180.3702	.1627	.0283	.6675	1.7959	.6858	.0599
201.3932	.1635	.0277	.6668	1.8103	.6832	.0539

MACH NO = 30.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8646	.0681	.8934	1.1566	-.0683	1.0120	1.0093
1.1214	.0735	.8567	.9004	.0551	.9904	.9858
1.5123	.0797	.8073	.7532	.2140	.9626	.9536
2.0307	.0849	.7479	.5772	.3924	.9313	.9141
2.8348	.0881	.6677	.4842	.6279	.8901	.8589
3.8369	.0878	.5845	.4288	.8752	.8469	.7987
4.8175	.0857	.5173	.4009	1.0787	.8112	.7475
5.1481	.0820	.4433	.3832	1.3038	.7719	.6877
7.6154	.0781	.3791	.3790	1.4942	.7386	.6319
8.9173	.0752	.3374	.3828	1.6221	.7162	.5895
10.5472	.0722	.2874	.3934	1.7395	.6956	.5437
11.9437	.0701	.2554	.4056	1.8099	.6833	.5099
13.6456	.0684	.2235	.4227	1.8665	.6734	.4739
16.7793	.0666	.1793	.4571	1.9193	.6657	.4194
19.5759	.0664	.1508	.4884	1.9048	.6667	.3804
22.5291	.0675	.1281	.5209	1.8691	.6730	.3463
25.0684	.0695	.1130	.5432	1.8211	.6813	.3216
27.6896	.0726	.1005	.5754	1.7605	.6920	.2995
29.9233	.0761	.0918	.5975	1.7037	.7019	.2829
32.2382	.0802	.0843	.6189	1.6437	.7124	.2676
34.2449	.0845	.0788	.6359	1.5936	.7212	.2556
36.3904	.0894	.0737	.6520	1.5444	.7298	.2439
38.5270	.0947	.0694	.6657	1.5023	.7371	.2333
40.5206	.0997	.0659	.6761	1.4708	.7426	.2241
42.8372	.1053	.0624	.6854	1.4446	.7472	.2144
45.1186	.1104	.0594	.6918	1.4295	.7499	.2056
47.7923	.1157	.0563	.6963	1.4235	.7509	.1962
50.4289	.1203	.0537	.6985	1.4269	.7503	.1877
53.6718	.1258	.0509	.6991	1.4389	.7482	.1782
57.0636	.1292	.0484	.6985	1.4560	.7452	.1692
61.4061	.1338	.0457	.6970	1.4796	.7411	.1590
65.7243	.1377	.0434	.6951	1.5030	.7370	.1500
71.0179	.1416	.0412	.6925	1.5313	.7321	.1403
77.1925	.1453	.0390	.6895	1.5626	.7266	.1304
83.8308	.1484	.0371	.6863	1.5937	.7211	.1212
91.7917	.1512	.0352	.6828	1.6272	.7153	.1118
99.5996	.1533	.0338	.6798	1.6557	.7103	.1038
108.8794	.1552	.0324	.6770	1.6841	.7053	.0958
117.9917	.1567	.0313	.6747	1.7074	.7012	.0890
128.7075	.1580	.0303	.6726	1.7302	.6973	.0821
139.7710	.1591	.0296	.6711	1.7484	.6941	.0763
151.8932	.1600	.0288	.6697	1.7660	.6910	.0704
164.1572	.1608	.0282	.6687	1.7798	.6886	.0655
178.7140	.1615	.0276	.6679	1.7931	.6862	.0604
200.6733	.1623	.0269	.6670	1.8084	.6836	.0542

MACH NO = 3.50 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8411	.0701	.9934	1.1889	-.0805	1.0159	1.0129
1.0876	.0782	.9496	.9300	.0375	.9921	.9855
1.3434	.0858	.9096	.7844	.1390	.9708	.9601
1.7514	.0962	.8517	.6628	.2723	.9428	.9221
2.2718	.1063	.7867	.5874	.4114	.9135	.8778
2.9172	.1149	.7175	.5408	.5549	.8834	.8285
3.6971	.1217	.6474	.5133	.6980	.8533	.7758
4.6186	.1271	.5795	.5002	.8328	.8249	.7216
5.6875	.1315	.5163	.4974	.9540	.7995	.6675
6.9090	.1350	.4593	.5012	1.0594	.7773	.6148
8.2885	.1380	.4092	.5092	1.1482	.7586	.5645
9.8315	.1407	.3660	.5199	1.2205	.7434	.5171
11.9076	.1435	.3227	.5342	1.2886	.7291	.4647
13.4333	.1453	.2985	.5439	1.3248	.7215	.4325
15.5064	.1473	.2729	.5557	1.3615	.7138	.3952
17.7029	.1487	.2556	.5646	1.3854	.7088	.3678
19.7287	.1505	.2374	.5750	1.4097	.7037	.3363
21.8197	.1517	.2252	.5827	1.4256	.7003	.3131
24.0504	.1530	.2147	.5898	1.4389	.6975	.2917
27.0443	.1544	.2036	.5980	1.4525	.6947	.2672
29.6118	.1555	.1962	.6039	1.4615	.6928	.2492
33.0481	.1567	.1884	.6107	1.4709	.6908	.2286
35.9882	.1577	.1832	.6156	1.4772	.6895	.2135
39.9156	.1588	.1776	.6211	1.4840	.6880	.1962
43.2707	.1597	.1739	.6251	1.4887	.6871	.1835
47.7473	.1606	.1699	.6296	1.4939	.6860	.1690
51.5684	.1614	.1672	.6329	1.4977	.6852	.1582
56.6647	.1622	.1644	.6365	1.5020	.6843	.1459
61.0144	.1628	.1624	.6392	1.5051	.6836	.1367
66.8178	.1636	.1603	.6422	1.5087	.6829	.1262
71.7741	.1641	.1589	.6444	1.5115	.6823	.1184
78.3924	.1647	.1574	.6468	1.5147	.6816	.1094
84.0534	.1651	.1564	.6486	1.5171	.6811	.1027
91.6136	.1656	.1552	.6506	1.5200	.6805	.0950
98.0860	.1660	.1545	.6521	1.5222	.6800	.0892
106.7456	.1664	.1536	.6538	1.5247	.6795	.0825
114.1615	.1666	.1531	.6550	1.5266	.6791	.0775
124.0896	.1670	.1525	.6564	1.5289	.6786	.0717
132.5957	.1672	.1520	.6575	1.5305	.6783	.0674
143.9864	.1675	.1516	.6587	1.5325	.6779	.0624
153.7477	.1677	.1513	.6596	1.5339	.6776	.0586
166.8210	.1679	.1509	.6606	1.5356	.6772	.0542
178.0250	.1680	.1507	.6614	1.5368	.6770	.0510
193.0311	.1682	.1505	.6623	1.5382	.6767	.0472
202.6025	.1683	.1503	.6628	1.5390	.6765	.0451

MACH NO = 5.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8533	.0693	.9436	1.1719	-.0742	1.0156	1.0109
1.0577	.0750	.9084	.9524	.0249	.9948	.9886
1.3338	.0815	.8656	.7842	.1383	.9709	.9610
1.6936	.0883	.8144	.6657	.2625	.9448	.9273
2.2688	.0959	.7421	.5712	.4272	.9102	.8781
3.0088	.1015	.6633	.5143	.6005	.8738	.8219
3.7255	.1045	.5992	.4880	.7382	.8448	.7740
4.7550	.1070	.5229	.4732	.8958	.8117	.7136
5.9688	.1089	.4535	.4729	1.0297	.7835	.6546
7.0492	.1104	.4040	.4800	1.1167	.7653	.6093
8.5518	.1125	.3497	.4949	1.2003	.7477	.5558
9.8823	.1147	.3122	.5098	1.2491	.7374	.5157
11.7070	.1178	.2723	.5298	1.2916	.7285	.4693
13.6823	.1215	.2398	.5493	1.3184	.7229	.4276
15.8023	.1255	.2134	.5672	1.3350	.7194	.3904
18.0640	.1296	.1921	.5829	1.3460	.7171	.3573
20.9633	.1344	.1717	.5985	1.3559	.7150	.3222
23.5334	.1383	.1582	.6090	1.3635	.7134	.2964
26.2440	.1419	.1473	.6176	1.3713	.7117	.2733
29.0964	.1453	.1382	.6246	1.3795	.7100	.2526
32.0931	.1483	.1308	.6303	1.3881	.7082	.2340
35.2379	.1511	.1247	.6349	1.3970	.7063	.2172
38.5371	.1535	.1196	.6387	1.4060	.7044	.2020
42.0008	.1557	.1153	.6419	1.4151	.7025	.1882
46.3955	.1580	.1110	.6450	1.4257	.7003	.1731
50.2792	.1598	.1080	.6471	1.4344	.6985	.1617
54.3888	.1613	.1055	.6489	1.4429	.6967	.1511
58.7481	.1626	.1033	.6505	1.4511	.6950	.1413
63.3807	.1638	.1014	.6518	1.4591	.6933	.1322
68.3093	.1649	.0998	.6530	1.4667	.6917	.1238
73.5561	.1658	.0984	.6541	1.4739	.6902	.1159
79.1437	.1667	.0972	.6550	1.4808	.6887	.1085
86.3323	.1675	.0959	.6561	1.4883	.6871	.1003
92.7552	.1682	.0951	.6569	1.4941	.6859	.0939
99.6000	.1687	.0943	.6577	1.4995	.6848	.0880
106.8952	.1693	.0936	.6584	1.5044	.6838	.0824
114.6710	.1697	.0931	.6591	1.5089	.6828	.0772
122.9593	.1701	.0926	.6598	1.5130	.6820	.0723
131.7939	.1704	.0921	.6604	1.5168	.6811	.0678
141.2108	.1707	.0917	.6610	1.5204	.6804	.0635
153.3338	.1710	.0914	.6617	1.5243	.6796	.0588
164.1702	.1712	.0911	.6622	1.5272	.6790	.0551
175.7210	.1714	.0908	.6627	1.5299	.6784	.0516
188.0335	.1715	.0906	.6632	1.5324	.6779	.0484
201.1583	.1716	.0904	.6637	1.5347	.6774	.0454

MACH NO = 10.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/R9
.8620	.0684	.9052	1.1601	-.0697	1.0146	1.0097
1.1289	.0744	.8613	.8976	.0567	.9881	.9813
1.4342	.0796	.8156	.7382	.1786	.9625	.9513
1.9221	.0851	.7497	.6077	.3420	.9281	.9071
2.5425	.0887	.6770	.5262	.5158	.8916	.8564
3.2908	.0899	.6028	.4758	.6920	.8545	.8024
4.1651	.0894	.5308	.4463	.8617	.8189	.7473
5.1565	.0879	.4643	.4318	1.0157	.7865	.6933
6.2517	.0862	.4050	.4289	1.1462	.7591	.6421
7.4352	.0846	.3535	.4347	1.2496	.7373	.5946
8.6925	.0835	.3097	.4465	1.3262	.7212	.5513
10.0110	.0830	.2726	.4621	1.3788	.7102	.5121
11.3608	.0831	.2358	.4839	1.4154	.7025	.4704
13.6658	.0843	.2015	.5110	1.4300	.6994	.4279
15.7492	.0865	.1745	.5379	1.4238	.7007	.3913
17.9075	.0897	.1529	.5634	1.4048	.7047	.3594
20.1392	.0938	.1356	.5867	1.3795	.7100	.3314
22.4424	.0985	.1217	.6071	1.3526	.7157	.3068
25.1588	.1045	.1089	.6269	1.3242	.7216	.2821
27.6102	.1100	.0998	.6410	1.3036	.7260	.2630
30.1433	.1157	.0924	.6523	1.2878	.7293	.2458
32.7903	.1214	.0861	.6614	1.2771	.7316	.2301
35.6064	.1270	.0807	.6685	1.2712	.7328	.2154
39.1209	.1334	.0754	.6744	1.2708	.7329	.1995
42.5375	.1387	.0713	.6776	1.2764	.7317	.1862
46.2499	.1436	.0677	.6793	1.2871	.7294	.1736
50.1852	.1478	.0646	.6798	1.3013	.7265	.1619
54.3776	.1515	.0620	.6794	1.3175	.7230	.1512
58.8575	.1548	.0598	.6785	1.3348	.7194	.1411
64.3627	.1579	.0576	.6773	1.3550	.7152	.1305
69.5397	.1603	.0560	.6759	1.3726	.7115	.1218
75.0722	.1622	.0545	.6745	1.3899	.7078	.1138
80.9819	.1639	.0533	.6731	1.4064	.7044	.1062
87.2945	.1653	.0522	.6717	1.4221	.7011	.0993
95.0373	.1665	.0512	.6703	1.4387	.6976	.0918
102.3068	.1674	.0504	.6692	1.4521	.6948	.0858
110.0702	.1681	.0497	.6682	1.4643	.6922	.0802
118.3609	.1687	.0491	.6674	1.4754	.6899	.0750
127.2157	.1692	.0486	.6668	1.4853	.6878	.0701
136.6746	.1696	.0482	.6663	1.4941	.6859	.0655
148.2791	.1699	.0477	.6658	1.5028	.6841	.0607
159.1798	.1702	.0474	.6656	1.5095	.6827	.0567
170.8296	.1704	.0471	.6655	1.5154	.6814	.0530
183.2829	.1705	.0468	.6654	1.5206	.6804	.0496
200.5690	.1707	.0465	.6654	1.5264	.6791	.0455

MACH NO = 15.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8636	.0683	.0979	1.1579	-.0588	1.0145	1.0094
1.1255	.0738	.0960	.8996	.0554	.9883	.9816
1.4263	.0787	.0900	.7403	.1763	.9629	.9521
1.9069	.0877	.0751	.6082	.3394	.9287	.9084
2.6469	.0970	.0594	.5126	.5473	.8850	.8485
3.3974	.0973	.0568	.4648	.7230	.8480	.7952
4.2646	.0960	.0572	.4358	.8921	.8125	.7415
5.2367	.0938	.0573	.4204	1.0460	.7801	.6893
6.5183	.0908	.0562	.4159	1.2007	.7476	.6307
7.6614	.0787	.0387	.4205	1.3015	.7264	.5863
8.8552	.0769	.0283	.4304	1.3773	.7105	.5461
10.0858	.0757	.0243	.4437	1.4310	.6992	.5101
11.5957	.0748	.0206	.4624	1.4707	.6908	.4719
13.8975	.0748	.0114	.4929	1.4923	.6863	.4235
16.2103	.0761	.0124	.5233	1.4836	.6881	.3840
18.2523	.0782	.0128	.5488	1.4609	.6929	.3548
20.5160	.0814	.0137	.5749	1.4265	.7001	.3271
22.7348	.0854	.0125	.5982	1.3895	.7081	.3040
24.9092	.0899	.0122	.6184	1.3509	.7160	.2842
27.0506	.0949	.0139	.6357	1.3164	.7233	.2671
29.1828	.1002	.0172	.6502	1.2864	.7296	.2520
31.3442	.1058	.0215	.6624	1.2615	.7348	.2384
33.5858	.1114	.0266	.6723	1.2421	.7389	.2257
35.9682	.1172	.0323	.6801	1.2290	.7417	.2136
38.5612	.1230	.0383	.6859	1.2226	.7430	.2019
41.4227	.1286	.0447	.6896	1.2237	.7428	.1903
44.2159	.1334	.0518	.6912	1.2305	.7413	.1803
47.7384	.1384	.0587	.6916	1.2439	.7385	.1690
51.7879	.1431	.0658	.6908	1.2623	.7347	.1576
56.5175	.1475	.0731	.6890	1.2847	.7299	.1462
61.8058	.1514	.0807	.6866	1.3092	.7248	.1352
67.5067	.1546	.0887	.6841	1.3338	.7196	.1251
73.6415	.1573	.0969	.6814	1.3575	.7146	.1157
80.2424	.1595	.1054	.6790	1.3799	.7099	.1071
87.3473	.1612	.1141	.6767	1.4008	.7055	.0992
94.9979	.1627	.1230	.6746	1.4199	.7015	.0919
103.2394	.1639	.1321	.6728	1.4372	.6979	.0851
111.0998	.1648	.1414	.6715	1.4510	.6950	.0795
120.5891	.1656	.1507	.6702	1.4649	.6921	.0737
130.8145	.1662	.1601	.6692	1.4770	.6895	.0683
141.8351	.1667	.1696	.6684	1.4876	.6873	.0633
153.7166	.1672	.1791	.6678	1.4969	.6853	.0586
166.5314	.1676	.1887	.6674	1.5049	.6837	.0543
180.3587	.1678	.1984	.6671	1.5119	.6822	.0504
200.5201	.1681	.2130	.6669	1.5197	.6805	.0455

MACH NO = 20.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8642	.0682	.8951	1.1572	-.0685	1.0144	1.0094
1.1242	.0736	.8526	.9005	.0549	.9885	.9818
1.4233	.0783	.8079	.7411	.1754	.9631	.9524
2.0109	.0840	.7296	.5886	.3721	.9218	.8995
2.6335	.0863	.6584	.5122	.5456	.8853	.8495
3.3752	.0864	.5865	.4636	.7212	.8484	.7967
4.4145	.0845	.5042	.4292	.9234	.8059	.7329
5.3895	.0819	.4421	.4149	1.0748	.7741	.6817
6.4449	.0793	.3873	.4106	1.2038	.7469	.6338
7.7914	.0764	.3316	.4150	1.3256	.7214	.5816
8.9605	.0744	.2929	.4241	1.4006	.7056	.5428
10.1576	.0730	.2603	.4365	1.4543	.6943	.5081
11.6151	.0718	.2280	.4539	1.4951	.6857	.4714
14.0552	.0713	.1868	.4855	1.5207	.6803	.4206
16.4609	.0721	.1573	.5168	1.5119	.6822	.3802
18.7915	.0741	.1358	.5459	1.4839	.6881	.3478
21.0239	.0769	.1198	.5721	1.4462	.6960	.3215
23.1526	.0805	.1077	.5955	1.4047	.7047	.3000
25.1868	.0847	.0983	.6159	1.3633	.7134	.2819
27.1474	.0893	.0918	.6337	1.3246	.7216	.2664
29.2551	.0947	.0842	.6503	1.2868	.7295	.2516
31.1689	.0999	.0792	.6630	1.2575	.7357	.2394
33.1281	.1053	.0748	.6736	1.2337	.7407	.2282
35.1859	.1108	.0709	.6822	1.2159	.7444	.2175
37.4044	.1163	.0673	.6887	1.2048	.7467	.2070
39.8129	.1217	.0640	.6932	1.2011	.7475	.1967
42.4327	.1268	.0610	.6957	1.2045	.7468	.1866
45.3568	.1317	.0581	.6965	1.2143	.7447	.1764
48.7018	.1364	.0553	.6961	1.2294	.7416	.1661
52.6034	.1409	.0527	.6947	1.2491	.7374	.1555
57.0253	.1450	.0502	.6925	1.2719	.7326	.1451
62.6351	.1491	.0477	.6894	1.2998	.7268	.1336
68.5968	.1524	.0456	.6862	1.3269	.7211	.1233
75.0754	.1552	.0438	.6831	1.3529	.7156	.1138
82.0660	.1574	.0422	.6803	1.3771	.7105	.1050
89.6193	.1593	.0409	.6777	1.3993	.7059	.0969
97.7869	.1608	.0398	.6755	1.4192	.7017	.0895
106.6284	.1620	.0388	.6736	1.4371	.6979	.0826
116.1709	.1631	.0380	.6721	1.4529	.6946	.0763
126.4943	.1639	.0373	.6708	1.4668	.6917	.0704
137.6555	.1646	.0367	.6698	1.4788	.6891	.0651
149.7287	.1652	.0362	.6690	1.4893	.6869	.0601
164.1622	.1657	.0358	.6684	1.4992	.6849	.0551
178.4314	.1661	.0354	.6680	1.5068	.6833	.0509
200.4395	.1665	.0350	.6678	1.5158	.6814	.0455

MACH NO = 25.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8645	.0682	.8941	1.1568	-.0684	1.0144	1.0093
1.1236	.0735	.8517	.9008	.0547	.9885	.9818
1.4219	.0782	.8071	.7415	.1750	.9632	.9525
2.0076	.0837	.7290	.5887	.3715	.9219	.8998
2.6272	.0860	.6581	.5121	.5448	.8855	.8500
3.3649	.0860	.5864	.4631	.7204	.8486	.7974
4.3974	.0840	.5044	.4281	.9229	.8060	.7339
5.3648	.0813	.4426	.4131	1.0751	.7740	.6829
6.4105	.0786	.3880	.4081	1.2053	.7466	.6353
7.7421	.0755	.3325	.4115	1.3289	.7206	.5834
8.8960	.0735	.2939	.4199	1.4058	.7045	.5448
10.0748	.0718	.2614	.4315	1.4615	.6928	.5104
11.5063	.0705	.2292	.4481	1.5049	.6837	.4740
14.1281	.0696	.1848	.4816	1.5353	.6773	.4192
16.6830	.0702	.1539	.5145	1.5260	.6792	.3768
19.1189	.0721	.1321	.5448	1.4955	.6856	.3437
21.2087	.0746	.1175	.5696	1.4585	.6934	.3195
23.3695	.0782	.1054	.5939	1.4138	.7028	.2979
25.4026	.0823	.0962	.6151	1.3691	.7122	.2801
27.3373	.0868	.0889	.6336	1.3272	.7210	.2650
29.2105	.0918	.0831	.6494	1.2895	.7289	.2519
31.0652	.0969	.0781	.6629	1.2569	.7358	.2401
32.9501	.1023	.0739	.6742	1.2300	.7414	.2292
34.9209	.1078	.0701	.6834	1.2096	.7457	.2188
37.0425	.1133	.0666	.6904	1.1962	.7486	.2086
39.3261	.1187	.0634	.6952	1.1906	.7497	.1987
41.8007	.1238	.0604	.6979	1.1926	.7493	.1889
44.2955	.1283	.0579	.6989	1.2003	.7477	.1800
47.4197	.1329	.0551	.6986	1.2143	.7447	.1699
51.0414	.1375	.0524	.6973	1.2330	.7408	.1596
55.0480	.1416	.0500	.6953	1.2544	.7363	.1496
59.5813	.1454	.0478	.6927	1.2783	.7313	.1396
64.9007	.1489	.0456	.6895	1.3047	.7257	.1295
71.3008	.1522	.0435	.6860	1.3334	.7197	.1191
78.4094	.1548	.0417	.6826	1.3610	.7139	.1094
86.1289	.1570	.0402	.6797	1.3861	.7086	.1005
94.5227	.1588	.0389	.6771	1.4085	.7039	.0923
103.6509	.1603	.0378	.6749	1.4284	.6997	.0848
112.6331	.1614	.0369	.6733	1.4445	.6964	.0785
123.3233	.1624	.0361	.6718	1.4600	.6931	.0721
134.9359	.1633	.0355	.6706	1.4735	.6903	.0663
147.5601	.1640	.0349	.6697	1.4851	.6878	.0609
161.2977	.1645	.0344	.6690	1.4950	.6857	.0560
176.2602	.1650	.0340	.6686	1.5035	.6840	.0515
200.4538	.1656	.0335	.6682	1.5136	.6818	.0455

MACH NO = 30.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8646	.0681	.8934	1.1566	-.0683	1.0144	1.0093
1.1233	.0735	.8511	.9010	.0546	.9885	.9819
1.4211	.0781	.8066	.7417	.1748	.9633	.9526
2.0057	.0836	.7287	.5888	.3711	.9220	.8999
2.6237	.0858	.6579	.5120	.5443	.8856	.8502
3.3592	.0858	.5864	.4627	.7199	.8487	.7978
4.3880	.0837	.5045	.4275	.9226	.8061	.7344
5.3512	.0810	.4428	.4122	1.0752	.7740	.6836
6.3915	.0782	.3883	.4067	1.2060	.7465	.6361
7.7150	.0751	.3329	.4097	1.3307	.7203	.5844
8.8605	.0729	.2944	.4176	1.4086	.7039	.5459
10.0294	.0712	.2620	.4288	1.4654	.6920	.5116
11.4467	.0698	.2298	.4450	1.5102	.6825	.4754
14.0355	.0687	.1855	.4776	1.5432	.6756	.4209
16.7698	.0692	.1525	.5126	1.5345	.6774	.3755
19.1388	.0708	.1313	.5421	1.5049	.6837	.3434
21.5463	.0737	.1148	.5708	1.4611	.6929	.3160
23.5946	.0770	.1036	.5940	1.4169	.7022	.2958
25.6805	.0813	.0944	.6162	1.3689	.7122	.2778
27.4883	.0856	.0877	.6339	1.3277	.7209	.2639
29.2326	.0902	.0823	.6493	1.2904	.7288	.2517
31.1081	.0955	.0774	.6635	1.2551	.7362	.2398
32.8482	.1006	.0734	.6745	1.2282	.7418	.2297
34.8279	.1062	.0696	.6842	1.2056	.7466	.2193
36.7774	.1115	.0664	.6911	1.1917	.7495	.2098
39.0526	.1170	.0631	.6963	1.1848	.7509	.1998
41.3028	.1218	.0603	.6990	1.1857	.7507	.1908
44.0334	.1268	.0574	.7002	1.1939	.7490	.1809
46.8985	.1312	.0549	.7000	1.2069	.7463	.1715
50.4790	.1359	.0522	.6986	1.2256	.7424	.1611
54.0538	.1397	.0499	.6968	1.2450	.7383	.1519
58.0400	.1433	.0478	.6945	1.2666	.7338	.1428
63.1018	.1469	.0456	.6914	1.2928	.7283	.1328
68.7057	.1501	.0436	.6881	1.3195	.7226	.1231
76.0254	.1531	.0416	.6842	1.3501	.7162	.1125
83.3705	.1554	.0400	.6811	1.3758	.7108	.1035
92.1057	.1575	.0385	.6782	1.4008	.7055	.0945
100.8359	.1591	.0374	.6759	1.4211	.7013	.0870
111.2062	.1605	.0363	.6739	1.4406	.6972	.0794
121.5475	.1615	.0355	.6724	1.4561	.6939	.0731
133.8261	.1625	.0348	.6711	1.4707	.6908	.0668
146.0871	.1632	.0342	.6701	1.4823	.6884	.0615
159.4104	.1638	.0337	.6694	1.4923	.6863	.0566
175.2800	.1643	.0332	.6689	1.5016	.6843	.0518
200.4194	.1650	.0327	.6685	1.5123	.6821	.0455

MACH NO = 3.50 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8411	.0701	.9934	1.1889	-.0805	1.0198	1.0129
1.0321	.0765	.9552	.9766	.0119	.9971	.9887
1.2724	.0836	.9116	.8213	.1092	.9732	.9607
1.6548	.0935	.8492	.6915	.2346	.9424	.9192
2.1416	.1030	.7802	.6111	.3629	.9109	.8713
2.7441	.1112	.7077	.5618	.4921	.8792	.8186
3.3147	.1165	.6497	.5376	.5933	.8543	.7742
4.1433	.1219	.5800	.5214	.7115	.8253	.7176
5.1039	.1264	.5157	.5168	.8158	.7997	.6616
6.1997	.1302	.4584	.5197	.9046	.7779	.6076
7.4335	.1336	.4085	.5271	.9779	.7599	.5563
8.5221	.1361	.3739	.5351	1.0259	.7481	.5178
10.0123	.1390	.3369	.5463	1.0743	.7362	.4730
11.6500	.1417	.3061	.5578	1.1125	.7268	.4319
13.4396	.1442	.2807	.5689	1.1427	.7194	.3945
15.3859	.1465	.2597	.5792	1.1669	.7134	.3605
17.4946	.1486	.2426	.5886	1.1865	.7086	.3297
19.7723	.1506	.2285	.5971	1.2025	.7047	.3019
22.2261	.1523	.2170	.6047	1.2157	.7015	.2767
24.3217	.1536	.2093	.6101	1.2247	.6992	.2583
27.1145	.1551	.2012	.6163	1.2344	.6969	.2373
30.1097	.1564	.1945	.6218	1.2426	.6949	.2183
33.3190	.1577	.1890	.6268	1.2496	.6931	.2010
36.7560	.1588	.1844	.6312	1.2558	.6916	.1853
40.4360	.1598	.1806	.6351	1.2613	.6903	.1710
44.3771	.1607	.1774	.6386	1.2661	.6891	.1579
48.5997	.1616	.1748	.6417	1.2706	.6880	.1459
52.1962	.1622	.1730	.6440	1.2738	.6872	.1371
56.9869	.1629	.1710	.6466	1.2776	.6863	.1269
62.1321	.1635	.1694	.6489	1.2811	.6854	.1175
67.6627	.1640	.1680	.6510	1.2844	.6846	.1088
73.6119	.1645	.1668	.6529	1.2874	.6839	.1008
80.0149	.1650	.1659	.6547	1.2901	.6832	.0934
86.9090	.1653	.1650	.6563	1.2926	.6826	.0865
94.3340	.1657	.1643	.6577	1.2949	.6820	.0802
102.3321	.1660	.1637	.6591	1.2970	.6815	.0744
109.1735	.1662	.1633	.6601	1.2986	.6811	.0700
118.3191	.1665	.1629	.6612	1.3004	.6807	.0649
128.1729	.1667	.1625	.6623	1.3020	.6803	.0602
138.7900	.1669	.1622	.6633	1.3035	.6799	.0558
150.2300	.1671	.1619	.6642	1.3049	.6796	.0517
162.5569	.1672	.1617	.6650	1.3061	.6793	.0480
175.8396	.1673	.1615	.6658	1.3073	.6790	.0445
190.1527	.1674	.1613	.6665	1.3083	.6787	.0413
202.3986	.1675	.1612	.6671	1.3091	.6785	.0389

MACH NO = 5.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8533	.0693	.9436	1.1719	-.0742	1.0182	1.0109
1.0584	.0749	.9040	.9540	.0240	.9941	.9855
1.3325	.0811	.8555	.7895	.1338	.9671	.9539
1.6859	.0875	.7990	.6745	.2513	.9383	.9160
2.2444	.0943	.7214	.5835	.4029	.9011	.8618
2.7957	.0984	.6562	.5381	.5258	.8709	.8143
3.6057	.1017	.5767	.5059	.6711	.8352	.7533
4.3503	.1036	.5170	.4945	.7750	.8097	.7048
5.4004	.1055	.4495	.4928	.8849	.7827	.6461
6.3346	.1071	.4017	.4986	.9552	.7654	.6015
7.6198	.1094	.3498	.5118	1.0217	.7491	.5494
8.7435	.1115	.3142	.5252	1.0599	.7397	.5106
10.2731	.1148	.2764	.5434	1.0929	.7316	.4660
12.3067	.1193	.2393	.5651	1.1170	.7257	.4174
14.1601	.1235	.2144	.5817	1.1289	.7228	.3812
16.5289	.1286	.1910	.5987	1.1382	.7205	.3431
18.6204	.1327	.1756	.6103	1.1443	.7190	.3154
21.2699	.1374	.1609	.6214	1.1515	.7172	.2860
23.5941	.1410	.1512	.6288	1.1579	.7156	.2644
26.5244	.1450	.1419	.6358	1.1664	.7136	.2415
29.6144	.1485	.1344	.6411	1.1754	.7114	.2212
32.3194	.1510	.1294	.6447	1.1832	.7095	.2061
35.7379	.1538	.1244	.6481	1.1926	.7071	.1897
38.7483	.1558	.1209	.6504	1.2006	.7052	.1772
42.5813	.1579	.1174	.6526	1.2100	.7029	.1636
46.6850	.1598	.1146	.6544	1.2193	.7006	.1511
50.3336	.1611	.1125	.6556	1.2268	.6987	.1415
55.0118	.1625	.1105	.6568	1.2356	.6966	.1309
59.1801	.1636	.1090	.6577	1.2425	.6949	.1227
64.5324	.1646	.1075	.6587	1.2504	.6929	.1135
69.3066	.1654	.1064	.6594	1.2565	.6914	.1064
75.4421	.1662	.1053	.6602	1.2632	.6898	.0985
82.0553	.1668	.1044	.6609	1.2694	.6883	.0912
87.9590	.1673	.1037	.6615	1.2742	.6871	.0856
95.5498	.1678	.1030	.6621	1.2794	.6858	.0793
102.3270	.1681	.1026	.6627	1.2834	.6848	.0744
111.0418	.1685	.1021	.6633	1.2877	.6838	.0689
118.8228	.1687	.1017	.6637	1.2910	.6830	.0646
128.8289	.1689	.1013	.6643	1.2946	.6821	.0599
139.6188	.1691	.1010	.6649	1.2978	.6813	.0555
149.2535	.1693	.1008	.6653	1.3001	.6807	.0521
161.6445	.1694	.1006	.6659	1.3026	.6801	.0482
172.7099	.1694	.1004	.6663	1.3044	.6797	.0453
186.9422	.1695	.1002	.6668	1.3064	.6792	.0419
202.2933	.1696	.1001	.6673	1.3080	.6788	.0389

MACH NO = 10.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8620	.0684	.9052	1.1601	-.0697	1.0171	1.0097
1.0642	.0729	.8670	.9484	.0271	.9934	.9848
1.3477	.0779	.8176	.7786	.1421	.9651	.9522
1.7929	.0831	.7483	.6408	.2914	.9284	.9051
2.3452	.0864	.6742	.5559	.4441	.8910	.8527
3.0060	.0877	.5994	.5032	.5955	.8538	.7976
3.7722	.0874	.5278	.4721	.7387	.8196	.7419
4.6348	.0862	.4622	.4564	.8665	.7872	.6878
5.5807	.0848	.4041	.4523	.9735	.7609	.6370
6.5952	.0837	.3541	.4568	1.0572	.7404	.5901
7.6642	.0829	.3116	.4672	1.1183	.7254	.5477
8.7756	.0827	.2758	.4813	1.1598	.7152	.5096
9.9201	.0830	.2459	.4977	1.1849	.7090	.4756
11.8044	.0845	.2077	.5256	1.1997	.7054	.4284
13.7392	.0873	.1786	.5531	1.1937	.7069	.3888
15.7146	.0910	.1563	.5786	1.1765	.7111	.3553
17.7245	.0956	.1389	.6012	1.1546	.7165	.3267
19.7664	.1007	.1253	.6206	1.1323	.7219	.3020
21.8434	.1063	.1144	.6368	1.1123	.7269	.2804
23.9689	.1120	.1056	.6500	1.0958	.7309	.2613
26.1708	.1179	.0983	.6607	1.0835	.7339	.2440
28.4941	.1238	.0922	.6691	1.0756	.7359	.2281
30.9969	.1297	.0869	.6755	1.0723	.7367	.2132
33.7460	.1354	.0823	.6799	1.0741	.7362	.1989
36.8155	.1408	.0782	.6825	1.0812	.7345	.1850
40.2885	.1459	.0745	.6835	1.0932	.7315	.1715
44.1628	.1503	.0713	.6833	1.1089	.7277	.1585
48.3396	.1541	.0687	.6822	1.1262	.7235	.1466
52.8492	.1573	.0664	.6807	1.1442	.7190	.1356
57.0885	.1596	.0647	.6791	1.1602	.7151	.1267
62.2904	.1618	.0630	.6773	1.1780	.7107	.1172
67.8996	.1635	.0616	.6754	1.1949	.7066	.1084
73.9471	.1649	.0604	.6737	1.2106	.7027	.1004
80.4663	.1660	.0594	.6722	1.2250	.6992	.0929
87.4936	.1668	.0586	.6710	1.2380	.6960	.0860
95.0687	.1675	.0578	.6699	1.2495	.6932	.0796
103.2355	.1680	.0572	.6691	1.2597	.6907	.0737
112.0417	.1684	.0567	.6684	1.2686	.6885	.0683
121.5387	.1687	.0562	.6680	1.2763	.6866	.0633
131.7827	.1689	.0559	.6677	1.2830	.6849	.0586
142.8347	.1691	.0555	.6675	1.2888	.6835	.0543
154.7610	.1692	.0553	.6675	1.2938	.6823	.0503
167.6336	.1692	.0550	.6675	1.2980	.6812	.0466
181.5299	.1693	.0548	.6677	1.3017	.6804	.0432
200.4682	.1693	.0546	.6679	1.3054	.6794	.0392

MACH NO = 15.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8636	.0683	.8979	1.1579	-.0688	1.0159	1.0094
1.1264	.0737	.8489	.9015	.0542	.9867	.9775
1.4198	.0781	.7990	.7476	.1692	.9585	.9442
1.8757	.0825	.7298	.6216	.3183	.9218	.8968
2.4342	.0850	.6568	.5428	.4702	.8845	.8449
3.0956	.0854	.5838	.4929	.6207	.8476	.7906
4.0170	.0840	.5014	.4581	.7898	.8060	.7257
4.8774	.0821	.4397	.4442	.9136	.7756	.6740
5.8053	.0801	.3857	.4404	1.0168	.7503	.6260
6.7844	.0783	.3394	.4442	1.0977	.7304	.5822
7.7999	.0770	.3002	.4531	1.1576	.7157	.5428
8.8395	.0761	.2674	.4654	1.1993	.7055	.5076
10.1054	.0757	.2348	.4829	1.2293	.6981	.4705
12.0159	.0762	.1971	.5111	1.2446	.6944	.4237
14.1211	.0781	.1665	.5419	1.2351	.6967	.3819
16.1788	.0812	.1443	.5701	1.2113	.7025	.3483
18.1721	.0851	.1278	.5949	1.1814	.7099	.3209
19.9100	.0892	.1165	.6143	1.1532	.7168	.3004
21.7901	.0942	.1065	.6329	1.1234	.7241	.2809
23.6404	.0996	.0986	.6486	1.0966	.7307	.2640
25.4951	.1052	.0921	.6617	1.0740	.7363	.2491
27.2033	.1105	.0872	.6715	1.0577	.7403	.2367
29.1906	.1164	.0824	.6802	1.0445	.7435	.2238
31.3280	.1223	.0781	.6868	1.0370	.7453	.2114
33.6791	.1282	.0742	.6912	1.0358	.7457	.1992
36.2724	.1337	.0707	.6937	1.0407	.7444	.1873
38.8795	.1384	.0678	.6944	1.0499	.7422	.1767
42.1860	.1433	.0647	.6939	1.0648	.7385	.1649
46.0580	.1479	.0619	.6922	1.0840	.7338	.1529
50.6063	.1520	.0592	.6896	1.1064	.7283	.1409
55.2959	.1553	.0571	.6869	1.1282	.7230	.1303
60.9615	.1582	.0550	.6838	1.1516	.7172	.1195
67.1116	.1605	.0533	.6809	1.1734	.7118	.1096
73.7900	.1624	.0519	.6783	1.1933	.7070	.1006
80.2933	.1637	.0508	.6763	1.2096	.7030	.0931
88.1160	.1649	.0498	.6744	1.2257	.6990	.0854
96.6225	.1659	.0489	.6728	1.2399	.6955	.0784
105.8738	.1666	.0482	.6716	1.2522	.6925	.0720
114.8920	.1671	.0477	.6707	1.2619	.6901	.0667
125.7492	.1675	.0472	.6700	1.2712	.6878	.0613
137.5655	.1678	.0468	.6694	1.2791	.6859	.0563
150.4339	.1681	.0464	.6691	1.2858	.6842	.0517
162.9934	.1682	.0462	.6690	1.2910	.6830	.0479
178.1335	.1684	.0459	.6689	1.2958	.6818	.0439
201.6381	.1685	.0456	.6690	1.3012	.6805	.0390

MACH NO = 20.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8642	.0682	.8951	1.1572	-.0685	1.0168	1.0094
1.1250	.0734	.8466	.9024	.0537	.9868	.9777
1.4165	.0778	.7970	.7486	.1682	.9587	.9446
1.8682	.0823	.7284	.6222	.3167	.9222	.8976
2.4208	.0843	.6560	.5427	.4683	.8850	.8460
3.0744	.0845	.5836	.4919	.6188	.8481	.7923
3.9832	.0870	.5017	.4561	.7885	.8064	.7279
4.8297	.0869	.4405	.4409	.9136	.7756	.6767
5.7407	.0786	.3868	.4358	1.0188	.7498	.6291
6.6982	.0766	.3408	.4381	1.1022	.7293	.5858
7.6881	.0750	.3019	.4457	1.1651	.7139	.5468
8.6976	.0738	.2692	.4568	1.2097	.7029	.5121
9.9209	.0730	.2368	.4729	1.2434	.6947	.4755
12.1566	.0731	.1923	.5054	1.2645	.6895	.4206
14.3334	.0747	.1614	.5373	1.2540	.6921	.3781
16.2281	.0771	.1412	.5636	1.2308	.6978	.3475
18.2035	.0806	.1249	.5892	1.1986	.7057	.3205
20.2689	.0848	.1127	.6114	1.1643	.7141	.2986
21.8438	.0895	.1034	.6306	1.1311	.7222	.2804
23.5577	.0945	.0961	.6470	1.1007	.7297	.2647
25.2464	.0999	.0900	.6609	1.0743	.7362	.2510
26.9498	.1054	.0850	.6724	1.0525	.7415	.2385
28.7128	.1111	.0806	.6818	1.0358	.7456	.2267
30.5881	.1168	.0767	.6890	1.0248	.7483	.2155
32.6243	.1224	.0731	.6942	1.0200	.7495	.2045
34.8299	.1278	.0698	.6973	1.0213	.7492	.1937
37.2763	.1329	.0667	.6987	1.0283	.7475	.1831
40.0624	.1378	.0638	.6985	1.0405	.7445	.1723
43.2869	.1424	.0610	.6972	1.0570	.7404	.1613
46.9060	.1466	.0585	.6951	1.0764	.7357	.1505
51.0520	.1503	.0561	.6923	1.0982	.7303	.1398
55.9620	.1537	.0539	.6890	1.1220	.7245	.1289
61.2716	.1565	.0520	.6858	1.1448	.7189	.1189
67.6943	.1590	.0502	.6826	1.1683	.7131	.1087
74.6937	.1611	.0487	.6798	1.1893	.7079	.0994
82.3299	.1627	.0474	.6774	1.2081	.7033	.0910
90.6649	.1640	.0464	.6754	1.2247	.6993	.0832
99.7616	.1650	.0455	.6737	1.2393	.6957	.0761
109.6891	.1658	.0448	.6724	1.2519	.6926	.0697
120.5268	.1664	.0442	.6714	1.2627	.6899	.0638
132.3643	.1669	.0437	.6707	1.2719	.6877	.0584
145.3016	.1672	.0433	.6702	1.2797	.6858	.0534
159.4477	.1675	.0430	.6699	1.2862	.6842	.0489
174.9197	.1678	.0427	.6698	1.2916	.6828	.0447
200.0524	.1680	.0424	.6698	1.2978	.6813	.0393

MACH NO = 25.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8645	.0682	.8941	1.1568	-.0684	1.0168	1.0093
1.1243	.0734	.8457	.9028	.0534	.9869	.9777
1.4149	.0776	.7963	.7490	.1677	.9588	.9448
1.8645	.0818	.7279	.6225	.3160	.9224	.8979
2.5365	.0842	.6412	.5306	.4977	.8778	.8360
3.2059	.0840	.5696	.4839	.6472	.8411	.7823
3.9671	.0825	.5021	.4551	.7879	.8065	.7290
4.8072	.0803	.4410	.4394	.9136	.7756	.6780
5.8962	.0775	.3777	.4334	1.0384	.7450	.6216
6.8512	.0755	.3333	.4363	1.1188	.7253	.5794
7.8337	.0738	.2958	.4441	1.1792	.7104	.5415
8.8316	.0726	.2642	.4550	1.2221	.6999	.5078
10.0358	.0717	.2330	.4708	1.2544	.6920	.4724
12.2227	.0716	.1901	.5024	1.2747	.6870	.4192
14.5203	.0730	.1580	.5359	1.2631	.6898	.3749
16.6813	.0757	.1359	.5658	1.2347	.6968	.3410
18.6910	.0793	.1202	.5919	1.1994	.7055	.3145
20.4111	.0832	.1095	.6128	1.1654	.7138	.2949
22.1815	.0880	.1005	.6325	1.1296	.7226	.2771
23.8772	.0931	.0935	.6493	1.0971	.7306	.2620
25.4011	.0981	.0882	.6624	1.0711	.7370	.2498
27.0708	.1037	.0833	.6743	1.0476	.7428	.2376
28.7958	.1094	.0791	.6839	1.0295	.7472	.2262
30.4740	.1147	.0756	.6908	1.0183	.7499	.2161
32.4454	.1204	.0721	.6962	1.0123	.7514	.2054
34.5708	.1258	.0689	.6994	1.0129	.7513	.1950
36.9295	.1309	.0659	.7008	1.0195	.7496	.1845
39.3834	.1354	.0632	.7007	1.0303	.7470	.1748
42.4271	.1400	.0605	.6994	1.0462	.7431	.1641
45.7809	.1442	.0579	.6974	1.0648	.7385	.1537
49.2532	.1476	.0558	.6949	1.0838	.7339	.1443
53.6820	.1511	.0535	.6917	1.1068	.7282	.1338
59.0746	.1544	.0513	.6881	1.1319	.7220	.1229
65.0402	.1571	.0495	.6847	1.1557	.7162	.1127
72.2061	.1595	.0477	.6815	1.1792	.7104	.1026
80.0671	.1614	.0463	.6788	1.2000	.7053	.0933
87.9456	.1628	.0452	.6767	1.2168	.7012	.0856
97.3390	.1640	.0443	.6748	1.2330	.6972	.0779
107.6419	.1650	.0435	.6733	1.2470	.6938	.0709
118.9471	.1657	.0428	.6721	1.2589	.6909	.0646
130.2816	.1663	.0423	.6713	1.2682	.6886	.0592
143.8175	.1667	.0419	.6707	1.2768	.6865	.0539
158.7010	.1671	.0415	.6704	1.2839	.6847	.0491
173.6456	.1674	.0412	.6702	1.2893	.6834	.0450
201.0931	.1678	.0408	.6703	1.2962	.6817	.0391

NSWC/WOL/TR 75-45

MACH NO = 30.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8646	.0681	.8934	1.1566	-.0683	1.0168	1.0093
1.1240	.0733	.8451	.9030	.0533	.9869	.9778
1.4139	.0776	.7958	.7493	.1675	.9589	.9449
1.8625	.0817	.7276	.6227	.3156	.9225	.8981
2.5327	.0840	.6411	.5305	.4972	.8779	.8363
3.1999	.0838	.5696	.4836	.6457	.8412	.7827
3.9583	.0823	.5022	.4546	.7875	.8056	.7295
4.7948	.0800	.4412	.4385	.9135	.7757	.6787
5.8783	.0772	.3790	.4321	1.0389	.7449	.6225
6.8277	.0750	.3337	.4347	1.1201	.7249	.5804
7.8035	.0733	.2962	.4421	1.1813	.7099	.5426
8.7936	.0720	.2647	.4527	1.2250	.6992	.5091
9.9869	.0711	.2336	.4680	1.2593	.6910	.4737
12.3423	.0708	.1875	.5019	1.2806	.6855	.4166
14.5950	.0721	.1565	.5347	1.2696	.6885	.3736
16.7010	.0746	.1351	.5639	1.2407	.6953	.3407
18.6486	.0780	.1198	.5895	1.2058	.7039	.3150
20.5967	.0827	.1078	.6135	1.1660	.7137	.2929
22.2772	.0869	.0994	.6326	1.1305	.7224	.2762
23.8812	.0918	.0928	.6490	1.0982	.7303	.2620
25.4462	.0970	.0874	.6629	1.0699	.7373	.2494
27.0116	.1023	.0828	.6745	1.0463	.7431	.2380
28.7587	.1082	.0785	.6847	1.0267	.7479	.2265
30.4726	.1137	.0749	.6920	1.0143	.7509	.2162
32.3089	.1191	.0716	.6972	1.0081	.7524	.2061
34.2756	.1242	.0686	.7004	1.0080	.7525	.1963
36.4424	.1291	.0657	.7019	1.0136	.7511	.1866
38.8957	.1338	.0630	.7019	1.0242	.7485	.1767
41.9055	.1385	.0602	.7007	1.0400	.7446	.1658
44.9306	.1425	.0578	.6988	1.0569	.7404	.1562
48.3196	.1461	.0556	.6964	1.0759	.7358	.1467
52.2513	.1494	.0535	.6934	1.0971	.7306	.1370
56.9717	.1526	.0514	.6901	1.1204	.7249	.1269
63.3515	.1557	.0492	.6861	1.1476	.7182	.1154
70.2209	.1583	.0474	.6828	1.1716	.7123	.1052
77.7459	.1603	.0459	.6800	1.1927	.7071	.0959
85.9956	.1620	.0447	.6776	1.2114	.7025	.0874
95.0327	.1633	.0437	.6756	1.2278	.6985	.0797
104.9277	.1644	.0428	.6740	1.2421	.6950	.0726
116.7151	.1653	.0421	.6727	1.2553	.6917	.0657
128.6885	.1659	.0415	.6718	1.2657	.6892	.0599
141.8273	.1664	.0411	.6711	1.2745	.6870	.0546
156.2529	.1668	.0407	.6707	1.2817	.6852	.0498
172.0943	.1672	.0404	.6705	1.2877	.6838	.0454
200.4127	.1676	.0400	.6706	1.2950	.6820	.0392

MACH NO = 3.50 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8411	.0701	.9934	1.1889	-.0805	1.0226	1.0129
1.0318	.0764	.9512	.9792	.0106	.9970	.9859
1.2706	.0834	.9029	.8273	.1047	.9706	.9543
1.5649	.0909	.8493	.7215	.2000	.9438	.9181
2.0210	.1000	.7769	.6355	.3193	.9102	.8671
2.5841	.1078	.7019	.5831	.4371	.8771	.8114
3.1162	.1129	.6426	.5573	.5276	.8517	.7650
3.8866	.1181	.5723	.5401	.6314	.8225	.7065
4.7763	.1226	.5083	.5351	.7207	.7974	.6491
5.5749	.1258	.4626	.5367	.7813	.7804	.6050
6.6825	.1294	.4127	.5431	.8441	.7627	.5530
7.6566	.1321	.3784	.5505	.8846	.7514	.5140
8.9853	.1354	.3420	.5611	.9249	.7400	.4690
10.4389	.1385	.3118	.5720	.9562	.7312	.4280
12.3506	.1419	.2828	.5843	.9853	.7230	.3839
14.0864	.1446	.2634	.5937	1.0044	.7177	.3510
15.9556	.1469	.2477	.6021	1.0201	.7133	.3214
17.9623	.1491	.2348	.6096	1.0333	.7096	.2947
20.1120	.1510	.2243	.6161	1.0445	.7064	.2706
22.8891	.1531	.2141	.6230	1.0559	.7032	.2447
25.3771	.1547	.2073	.6280	1.0640	.7009	.2254
28.0328	.1560	.2017	.6325	1.0712	.6989	.2079
30.8671	.1573	.1971	.6364	1.0775	.6971	.1920
33.8932	.1584	.1932	.6400	1.0832	.6955	.1775
37.7993	.1596	.1894	.6437	1.0893	.6938	.1618
41.3043	.1604	.1868	.6465	1.0939	.6925	.1498
45.0579	.1612	.1846	.6490	1.0982	.6913	.1389
49.0823	.1619	.1828	.6512	1.1021	.6902	.1287
54.3022	.1626	.1809	.6536	1.1063	.6890	.1176
59.0071	.1631	.1796	.6554	1.1095	.6881	.1091
64.0628	.1636	.1786	.6571	1.1125	.6873	.1013
69.4975	.1640	.1777	.6587	1.1152	.6865	.0940
75.3408	.1643	.1769	.6601	1.1176	.6859	.0873
82.9371	.1647	.1761	.6616	1.1202	.6851	.0798
89.7945	.1650	.1756	.6628	1.1222	.6846	.0741
97.1705	.1652	.1751	.6639	1.1240	.6841	.0688
105.1048	.1654	.1747	.6650	1.1257	.6836	.0639
113.6403	.1656	.1744	.6659	1.1272	.6832	.0594
124.7413	.1658	.1741	.6670	1.1288	.6827	.0543
134.7659	.1659	.1739	.6678	1.1300	.6824	.0505
145.5512	.1660	.1737	.6686	1.1311	.6821	.0469
157.1553	.1661	.1735	.6693	1.1321	.6818	.0436
169.6406	.1662	.1734	.6699	1.1330	.6815	.0405
185.8813	.1662	.1732	.6707	1.1339	.6813	.0370
200.5495	.1663	.1731	.6712	1.1346	.6811	.0344

MACH NO = 5.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8533	.0693	.9436	1.1719	-.0742	1.0208	1.0109
1.0583	.0747	.8993	.9564	.0227	.9936	.9823
1.2554	.0792	.8599	.8307	.1016	.9714	.9563
1.6761	.0867	.7848	.6842	.2395	.9327	.9051
2.0965	.0919	.7201	.6109	.3504	.9015	.8592
2.7235	.0965	.6388	.5546	.4844	.8638	.7987
3.3062	.0989	.5763	.5284	.5839	.8359	.7497
4.1351	.1012	.5040	.5139	.6928	.8053	.6894
4.8761	.1028	.4520	.5119	.7655	.7848	.6433
5.8960	.1048	.3950	.5179	.8371	.7647	.5890
6.7845	.1067	.3556	.5272	.8799	.7527	.5486
7.7356	.1090	.3214	.5389	.9115	.7438	.5111
9.0168	.1122	.2852	.5551	.9387	.7361	.4681
10.6905	.1156	.2498	.5747	.9585	.7306	.4216
12.5251	.1215	.2215	.5928	.9698	.7274	.3803
14.5414	.1267	.1990	.6086	.9770	.7254	.3433
16.7106	.1318	.1814	.6214	.9830	.7237	.3108
18.9965	.1365	.1678	.6315	.9891	.7220	.2826
21.3995	.1409	.1572	.6392	.9959	.7201	.2579
23.9232	.1448	.1488	.6450	1.0034	.7179	.2363
26.5767	.1492	.1422	.6495	1.0115	.7157	.2172
29.3762	.1512	.1368	.6529	1.0200	.7133	.2001
32.3442	.1538	.1325	.6556	1.0287	.7109	.1847
35.5059	.1561	.1289	.6576	1.0374	.7084	.1707
38.8859	.1581	.1259	.6592	1.0462	.7059	.1579
42.5078	.1598	.1235	.6604	1.0548	.7035	.1461
46.3948	.1612	.1214	.6613	1.0631	.7012	.1353
50.5709	.1624	.1197	.6621	1.0711	.6989	.1254
55.0615	.1635	.1182	.6628	1.0785	.6969	.1162
59.8934	.1643	.1170	.6634	1.0854	.6949	.1077
65.0944	.1650	.1160	.6640	1.0918	.6931	.0998
70.6944	.1656	.1151	.6645	1.0976	.6915	.0926
76.7248	.1661	.1143	.6650	1.1030	.6900	.0858
83.2193	.1665	.1137	.6655	1.1078	.6886	.0796
90.2141	.1668	.1132	.6660	1.1121	.6874	.0738
97.7480	.1671	.1127	.6665	1.1160	.6863	.0685
105.8630	.1673	.1124	.6670	1.1194	.6854	.0635
114.6043	.1674	.1120	.6675	1.1224	.6845	.0589
124.0206	.1675	.1118	.6680	1.1250	.6838	.0546
134.1643	.1676	.1115	.6685	1.1273	.6831	.0507
145.0925	.1677	.1113	.6690	1.1293	.6826	.0470
156.8664	.1677	.1112	.6696	1.1310	.6821	.0436
169.5523	.1677	.1110	.6701	1.1325	.6817	.0405
183.2220	.1677	.1109	.6705	1.1338	.6813	.0376
200.5163	.1677	.1108	.6711	1.1351	.6809	.0344

MACH NO = 10.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0684	.9055	1.1617	-.0703	1.0198	1.0098
1.0620	.0727	.8626	.9523	.0249	.9930	.9818
1.3325	.0773	.8098	.7900	.1324	.9628	.9465
1.7485	.0819	.7380	.6580	.2682	.9246	.8968
2.2551	.0849	.6632	.5761	.4030	.8867	.8430
2.8535	.0860	.5891	.5247	.5338	.8500	.7872
3.5391	.0858	.5191	.4944	.6546	.8160	.7317
4.3026	.0849	.4557	.4790	.7604	.7863	.6784
5.1314	.0838	.4001	.4748	.8473	.7619	.6287
6.0118	.0830	.3524	.4787	.9141	.7431	.5834
6.9309	.0827	.3121	.4883	.9621	.7296	.5425
7.6868	.0827	.2846	.4984	.9888	.7221	.5129
8.6506	.0832	.2553	.5131	1.0101	.7161	.4796
10.4210	.0853	.2139	.5415	1.0235	.7123	.4285
12.2184	.0887	.1834	.5694	1.0170	.7141	.3866
13.8278	.0926	.1629	.5920	1.0030	.7181	.3555
15.6446	.0977	.1452	.6141	.9839	.7234	.3259
17.4684	.1033	.1315	.6326	.9652	.7287	.3008
19.1042	.1086	.1219	.6463	.9505	.7328	.2814
20.9825	.1147	.1131	.6589	.9372	.7366	.2619
22.9367	.1209	.1059	.6690	.9276	.7393	.2443
24.7767	.1264	.1004	.6759	.9226	.7407	.2298
27.0149	.1325	.0951	.6817	.9214	.7410	.2143
29.4952	.1383	.0905	.6854	.9251	.7400	.1994
31.9652	.1432	.0868	.6872	.9326	.7379	.1865
35.1192	.1483	.0830	.6875	.9452	.7343	.1723
38.7612	.1529	.0797	.6866	.9614	.7298	.1583
42.3962	.1563	.0772	.6851	.9778	.7252	.1465
46.8307	.1594	.0748	.6829	.9967	.7199	.1342
51.6594	.1619	.0728	.6805	1.0151	.7147	.1230
56.3084	.1636	.0714	.6785	1.0307	.7103	.1139
61.9721	.1650	.0700	.6764	1.0468	.7058	.1044
68.1331	.1662	.0688	.6746	1.0613	.7017	.0957
74.0622	.1669	.0680	.6733	1.0727	.6985	.0887
81.2854	.1675	.0672	.6721	1.0841	.6953	.0813
89.1456	.1680	.0665	.6712	1.0939	.6925	.0746
96.7137	.1682	.0660	.6706	1.1014	.6904	.0691
105.9390	.1684	.0656	.6702	1.1087	.6884	.0635
115.9839	.1686	.0652	.6699	1.1149	.6866	.0582
125.6613	.1686	.0649	.6698	1.1195	.6853	.0540
137.4651	.1686	.0646	.6699	1.1239	.6841	.0495
150.3251	.1686	.0644	.6700	1.1275	.6831	.0455
162.7208	.1685	.0643	.6703	1.1301	.6823	.0421
177.8460	.1685	.0641	.6706	1.1325	.6817	.0387
200.1471	.1684	.0639	.6712	1.1348	.6810	.0345

MACH NO = 15.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0682	.8985	1.1617	-.0703	1.0198	1.0098
1.1184	.0733	.8443	.9098	.0491	.9862	.9742
1.3954	.0774	.7916	.7624	.1556	.9563	.9386
1.8177	.0813	.7204	.6409	.2902	.9184	.8891
2.3274	.0834	.6469	.5641	.4240	.8808	.8358
2.9238	.0838	.5747	.5152	.5536	.8444	.7811
3.6004	.0829	.5070	.4855	.6734	.8107	.7271
4.3461	.0814	.4460	.4696	.7787	.7811	.6756
5.1467	.0798	.3926	.4641	.8660	.7566	.6279
5.9875	.0784	.3470	.4662	.9341	.7374	.5845
6.8553	.0774	.3084	.4737	.9842	.7233	.5456
7.7391	.0768	.2760	.4847	1.0189	.7136	.5110
8.8095	.0767	.2439	.5006	1.0430	.7066	.4745
10.5907	.0779	.2033	.5297	1.0564	.7031	.4241
12.5115	.0806	.1717	.5607	1.0459	.7060	.3806
14.1990	.0840	.1510	.5860	1.0250	.7116	.3491
15.9802	.0886	.1343	.6101	1.0002	.7189	.3210
17.5368	.0933	.1228	.6288	.9763	.7256	.2999
19.2001	.0989	.1129	.6462	.9520	.7324	.2803
20.6934	.1042	.1058	.6595	.9326	.7379	.2647
22.3508	.1103	.0993	.6716	.9151	.7428	.2493
23.9062	.1159	.0943	.6805	.9032	.7461	.2365
25.7166	.1220	.0895	.6879	.8949	.7485	.2230
27.5005	.1276	.0856	.6928	.8922	.7492	.2112
29.6549	.1335	.0817	.6961	.8945	.7486	.1985
31.8239	.1385	.0785	.6974	.9013	.7467	.1872
34.5317	.1437	.0752	.6973	.9133	.7433	.1748
37.3831	.1481	.0724	.6960	.9281	.7391	.1633
41.0060	.1524	.0695	.6935	.9474	.7337	.1508
44.8241	.1558	.0671	.6907	.9670	.7282	.1395
49.8127	.1591	.0647	.6872	.9900	.7217	.1271
54.9025	.1615	.0629	.6841	1.0101	.7161	.1165
61.0093	.1636	.0612	.6811	1.0301	.7105	.1059
67.0678	.1650	.0599	.6788	1.0463	.7059	.0971
74.3464	.1662	.0588	.6766	1.0621	.7015	.0884
81.5757	.1671	.0580	.6750	1.0747	.6979	.0811
90.2673	.1677	.0572	.6736	1.0867	.6945	.0738
98.9042	.1681	.0567	.6727	1.0961	.6919	.0677
109.2934	.1685	.0562	.6719	1.1047	.6895	.0616
119.6236	.1686	.0558	.6715	1.1113	.6876	.0566
132.0576	.1688	.0554	.6712	1.1174	.6859	.0515
145.7246	.1688	.0552	.6712	1.1223	.6846	.0468
159.3238	.1689	.0550	.6713	1.1259	.6835	.0430
175.7004	.1689	.0548	.6715	1.1290	.6827	.0391
200.6859	.1689	.0546	.6720	1.1322	.6818	.0344

MACH NO = 20.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0681	.8959	1.1617	-.0703	1.0198	1.0000
1.1162	.0731	.8422	.9112	.0483	.9864	.9745
1.3906	.0770	.7899	.7639	.1542	.9567	.9392
1.8087	.0808	.7194	.6419	.2882	.9190	.8901
2.3129	.0828	.6464	.5645	.4217	.8815	.8373
2.9020	.0830	.5747	.5147	.5513	.8450	.7830
3.7112	.0817	.4948	.4796	.6943	.8049	.7190
4.4570	.0799	.4356	.4648	.7974	.7759	.6686
5.2519	.0781	.3840	.4598	.8826	.7519	.6221
6.0811	.0766	.3430	.4619	.9491	.7332	.5801
6.9317	.0754	.3029	.4691	.9982	.7194	.5425
7.7933	.0746	.2717	.4795	1.0324	.7098	.5090
8.8306	.0743	.2409	.4947	1.0573	.7028	.4739
10.7102	.0751	.1984	.5252	1.0708	.6990	.4211
12.6847	.0775	.1666	.5573	1.0587	.7024	.3771
14.5461	.0811	.1448	.5856	1.0344	.7092	.3432
16.1520	.0851	.1300	.6080	1.0083	.7166	.3185
17.8019	.0900	.1181	.6291	.9794	.7247	.2966
19.3757	.0954	.1091	.6469	.9523	.7323	.2784
20.7813	.1006	.1024	.6606	.9304	.7385	.2639
22.3100	.1064	.0965	.6731	.9104	.7441	.2497
23.8735	.1124	.0914	.6832	.8952	.7484	.2367
25.3771	.1179	.0874	.6904	.8856	.7511	.2254
27.1434	.1239	.0834	.6961	.8805	.7525	.2135
29.0563	.1296	.0798	.6997	.8809	.7524	.2019
30.9813	.1346	.0768	.7013	.8860	.7510	.1914
33.3484	.1397	.0736	.7014	.8962	.7481	.1800
36.0655	.1444	.0707	.7001	.9107	.7440	.1684
38.8310	.1483	.0682	.6981	.9264	.7396	.1581
42.2636	.1521	.0657	.6953	.9457	.7342	.1469
46.3222	.1556	.0633	.6919	.9669	.7282	.1355
50.8178	.1584	.0612	.6886	.9876	.7224	.1248
56.7038	.1612	.0591	.6850	1.0105	.7160	.1131
63.2244	.1633	.0575	.6819	1.0309	.7102	.1025
69.7758	.1649	.0562	.6795	1.0474	.7056	.0937
77.6167	.1662	.0551	.6773	1.0634	.7011	.0849
86.2538	.1671	.0542	.6755	1.0773	.6972	.0770
94.9403	.1677	.0536	.6742	1.0882	.6941	.0704
105.3423	.1682	.0530	.6732	1.0982	.6913	.0638
116.8106	.1685	.0525	.6725	1.1065	.6890	.0579
128.3592	.1687	.0522	.6721	1.1128	.6872	.0529
142.2051	.1689	.0518	.6719	1.1185	.6856	.0480
157.4854	.1690	.0516	.6720	1.1230	.6844	.0435
172.8799	.1691	.0514	.6721	1.1262	.6834	.0397
201.2670	.1692	.0512	.6726	1.1301	.6823	.0343

MACH NO = 25.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RR
.8608	.0681	.8948	1.1617	-.0703	1.0198	1.0098
1.1151	.0730	.8414	.9119	.0479	.9865	.9747
1.3884	.0769	.7893	.7647	.1535	.9569	.9395
1.8045	.0806	.7190	.6424	.2873	.9192	.8906
2.3061	.0825	.6463	.5646	.4207	.8818	.8379
2.8919	.0827	.5749	.5144	.5503	.8453	.7839
3.6958	.0813	.4951	.4788	.6936	.8051	.7201
4.4358	.0794	.4361	.4634	.7972	.7759	.6699
5.2238	.0775	.3847	.4578	.8833	.7517	.6237
6.0447	.0758	.3408	.4593	.9507	.7328	.5818
6.8855	.0746	.3037	.4659	1.0010	.7186	.5444
7.7358	.0737	.2726	.4758	1.0363	.7087	.5111
8.7576	.0732	.2418	.4904	1.0625	.7013	.4762
10.7672	.0738	.1902	.5229	1.0781	.6970	.4197
12.6853	.0759	.1654	.5542	1.0663	.7003	.3771
14.6244	.0794	.1425	.5839	1.0404	.7076	.3419
16.2817	.0835	.1276	.6075	1.0120	.7156	.3167
17.9538	.0885	.1158	.6294	.9807	.7243	.2948
19.4138	.0935	.1075	.6466	.9536	.7320	.2780
20.8266	.0988	.1009	.6611	.9295	.7387	.2634
22.3419	.1048	.0950	.6742	.9078	.7448	.2494
23.7641	.1104	.0904	.6840	.8922	.7492	.2376
25.3774	.1165	.0860	.6922	.8805	.7525	.2254
26.9775	.1221	.0824	.6977	.8748	.7541	.2146
28.6887	.1274	.0791	.7013	.8743	.7543	.2040
30.7211	.1329	.0758	.7032	.8791	.7529	.1928
32.8357	.1377	.0729	.7033	.8882	.7504	.1823
35.4225	.1425	.0700	.7022	.9021	.7464	.1710
38.0375	.1464	.0675	.7002	.9174	.7421	.1609
40.9746	.1500	.0652	.6977	.9345	.7373	.1509
44.6897	.1535	.0628	.6944	.9551	.7316	.1399
48.8240	.1565	.0606	.6910	.9757	.7258	.1293
54.3554	.1595	.0584	.6872	.9991	.7192	.1175
60.2706	.1619	.0567	.6840	1.0195	.7134	.1071
66.9464	.1638	.0552	.6813	1.0381	.7082	.0973
74.9396	.1654	.0539	.6787	1.0559	.7032	.0877
83.0793	.1665	.0530	.6768	1.0702	.6992	.0797
92.8074	.1674	.0521	.6751	1.0836	.6954	.0719
102.7164	.1679	.0515	.6740	1.0940	.6925	.0653
113.6147	.1683	.0510	.6732	1.1027	.6901	.0594
126.6601	.1687	.0506	.6726	1.1104	.6879	.0536
139.9688	.1689	.0503	.6724	1.1162	.6863	.0487
154.6208	.1691	.0500	.6724	1.1208	.6850	.0442
172.1664	.1692	.0498	.6725	1.1247	.6839	.0399
201.3251	.1694	.0495	.6730	1.1288	.6827	.0343

MACH NO = 30.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8608	.0681	.8942	1.1617	-.0703	1.0198	1.0098
1.1145	.0730	.8409	.9123	.0476	.9866	.9747
1.3872	.0768	.7889	.7651	.1531	.9570	.9396
1.8022	.0804	.7188	.6427	.2858	.9194	.8908
2.4126	.0825	.6317	.5527	.4465	.8745	.8276
3.0126	.0823	.5611	.5067	.5748	.8384	.7736
3.6873	.0810	.4953	.4783	.6932	.8052	.7207
4.4242	.0791	.4363	.4627	.7972	.7759	.6707
5.3695	.0768	.3756	.4564	.8987	.7474	.6158
6.1906	.0751	.3332	.4588	.9631	.7293	.5749
7.0286	.0739	.2975	.4659	1.0108	.7159	.5385
7.8735	.0730	.2674	.4760	1.0440	.7065	.5061
8.8859	.0725	.2377	.4906	1.0684	.6997	.4721
11.0299	.0732	.1907	.5254	1.0818	.6959	.4133
12.9023	.0754	.1618	.5558	1.0682	.6997	.3728
14.7835	.0788	.1402	.5846	1.0419	.7071	.3393
16.5119	.0830	.1251	.6093	1.0111	.7158	.3135
18.1121	.0878	.1141	.6305	.9799	.7246	.2928
19.6199	.0931	.1057	.6485	.9509	.7327	.2757
21.0739	.0987	.0991	.6635	.9254	.7399	.2610
22.5127	.1044	.0936	.6760	.9042	.7458	.2479
23.8629	.1098	.0894	.6853	.8890	.7501	.2368
25.3930	.1157	.0852	.6933	.8775	.7534	.2253
27.0336	.1215	.0815	.6990	.8714	.7551	.2142
28.7881	.1271	.0782	.7026	.8709	.7552	.2035
30.7141	.1323	.0751	.7043	.8758	.7538	.1928
32.8903	.1372	.0721	.7043	.8854	.7511	.1821
35.1158	.1414	.0696	.7032	.8976	.7477	.1723
37.7274	.1455	.0670	.7013	.9131	.7433	.1620
40.6617	.1492	.0647	.6987	.9305	.7385	.1519
44.0764	.1525	.0624	.6955	.9499	.7330	.1416
48.1879	.1557	.0602	.6920	.9710	.7271	.1309
53.2485	.1586	.0580	.6884	.9932	.7208	.1197
59.0093	.1611	.0562	.6851	1.0140	.7150	.1091
65.0294	.1630	.0548	.6824	1.0316	.7100	.0999
72.6206	.1648	.0534	.6798	1.0497	.7050	.0903
81.0886	.1661	.0523	.6776	1.0656	.7005	.0815
90.4518	.1671	.0514	.6759	1.0793	.6966	.0736
100.8085	.1677	.0508	.6745	1.0910	.6933	.0665
112.2718	.1682	.0502	.6736	1.1007	.6906	.0601
124.9692	.1686	.0498	.6730	1.1086	.6884	.0543
137.9046	.1689	.0495	.6727	1.1145	.6867	.0494
153.3732	.1691	.0492	.6726	1.1196	.6853	.0446
170.5131	.1693	.0489	.6727	1.1236	.6842	.0403
200.5605	.1696	.0486	.6732	1.1278	.6830	.0344

MACH NO = 3.50 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8411	.0701	.9934	1.1889	-.0805	1.0255	1.0129
1.0315	.0763	.9467	.9821	.0091	.9971	.9828
1.2045	.0814	.9078	.8660	.0773	.9755	.9571
1.5591	.0904	.8369	.7307	.1907	.9396	.9082
1.9089	.0974	.7764	.6609	.2799	.9113	.8647
2.4360	.1049	.6994	.6049	.3881	.8771	.8065
2.9329	.1098	.6392	.5773	.4701	.8511	.7584
3.6504	.1149	.5686	.5587	.5624	.8219	.6982
4.3021	.1185	.5171	.5533	.6258	.8018	.6513
5.2137	.1227	.4602	.5540	.6921	.7808	.5953
6.0203	.1257	.4207	.5584	.7354	.7671	.5532
6.8953	.1286	.3864	.5649	.7707	.7559	.5138
8.0846	.1321	.3502	.5747	.8054	.7449	.4685
9.3799	.1355	.3205	.5849	.8322	.7364	.4274
11.0743	.1393	.2920	.5964	.8569	.7286	.3834
12.9233	.1428	.2699	.6067	.8760	.7225	.3447
14.5845	.1454	.2552	.6142	.8891	.7184	.3161
16.7269	.1482	.2412	.6220	.9023	.7142	.2854
18.6415	.1502	.2318	.6276	.9118	.7112	.2627
21.1030	.1524	.2228	.6334	.9217	.7080	.2383
23.2992	.1540	.2168	.6376	.9290	.7057	.2201
26.1225	.1556	.2109	.6420	.9368	.7033	.2003
28.6448	.1568	.2069	.6452	.9426	.7014	.1855
31.8946	.1581	.2030	.6486	.9489	.6994	.1693
34.8064	.1591	.2003	.6511	.9537	.6979	.1571
38.5701	.1600	.1977	.6537	.9590	.6962	.1436
42.6572	.1608	.1955	.6561	.9638	.6947	.1314
46.3338	.1614	.1940	.6579	.9674	.6936	.1221
51.1007	.1621	.1925	.6598	.9713	.6923	.1118
55.3937	.1625	.1914	.6613	.9743	.6914	.1039
60.9642	.1630	.1904	.6630	.9775	.6903	.0951
65.9841	.1633	.1897	.6642	.9799	.6896	.0885
72.5006	.1636	.1890	.6656	.9825	.6888	.0811
78.3750	.1639	.1885	.6667	.9845	.6881	.0754
86.0028	.1641	.1880	.6679	.9866	.6875	.0691
92.8804	.1642	.1876	.6688	.9881	.6870	.0642
101.8123	.1644	.1873	.6699	.9898	.6865	.0589
111.5480	.1645	.1870	.6708	.9913	.6860	.0540
120.3276	.1646	.1868	.6716	.9924	.6856	.0502
131.7312	.1647	.1866	.6724	.9935	.6853	.0460
142.0156	.1648	.1864	.6731	.9944	.6850	.0428
155.3750	.1648	.1863	.6738	.9953	.6847	.0393
167.4242	.1648	.1862	.6744	.9959	.6845	.0365
183.0771	.1649	.1861	.6750	.9966	.6843	.0335
200.1441	.1649	.1860	.6756	.9972	.6841	.0307

MACH NO = 5.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0690	.9460	1.1854	-.0792	1.0251	1.0125
1.0446	.0743	.8974	.9700	.0154	.9951	.9808
1.2372	.0787	.8549	.8448	.0914	.9710	.9524
1.6416	.0857	.7760	.7003	.2208	.9301	.8976
2.0317	.0904	.7109	.6294	.3202	.8986	.8504
2.6044	.0946	.6307	.5744	.4376	.8614	.7895
3.1311	.0969	.5699	.5490	.5230	.8343	.7407
3.8726	.0992	.5002	.5338	.6151	.8051	.6815
4.5290	.1009	.4505	.5311	.6757	.7860	.6364
5.4237	.1030	.3963	.5358	.7349	.7672	.5837
6.1954	.1050	.3590	.5438	.7700	.7561	.5449
7.0148	.1073	.3267	.5541	.7958	.7479	.5089
8.1043	.1105	.2926	.5685	.8179	.7409	.4678
9.5082	.1149	.2592	.5860	.8340	.7358	.4237
11.2866	.1205	.2287	.6048	.8443	.7325	.3785
12.9433	.1255	.2083	.6187	.8498	.7308	.3443
15.0500	.1313	.1894	.6320	.8553	.7291	.3089
17.0191	.1361	.1767	.6410	.8607	.7274	.2817
19.0972	.1405	.1667	.6479	.8669	.7254	.2578
21.6461	.1450	.1576	.6538	.8750	.7228	.2335
23.9513	.1484	.1516	.6575	.8825	.7204	.2152
26.8068	.1518	.1459	.6607	.8918	.7175	.1961
29.4203	.1543	.1421	.6626	.9000	.7149	.1814
32.6909	.1568	.1384	.6643	.9098	.7118	.1658
35.7060	.1586	.1358	.6652	.9181	.7092	.1536
38.9369	.1601	.1337	.6660	.9262	.7066	.1424
43.0057	.1615	.1316	.6666	.9352	.7038	.1305
46.7737	.1626	.1301	.6670	.9425	.7015	.1210
51.5277	.1635	.1287	.6675	.9503	.6990	.1109
55.9359	.1642	.1277	.6678	.9565	.6970	.1030
61.5019	.1648	.1267	.6682	.9630	.6950	.0944
66.6653	.1652	.1260	.6686	.9680	.6934	.0876
72.2206	.1655	.1254	.6689	.9725	.6919	.0813
79.2375	.1658	.1249	.6694	.9771	.6905	.0746
85.7485	.1659	.1245	.6699	.9806	.6894	.0693
93.9734	.1661	.1240	.6704	.9841	.6883	.0635
101.6060	.1662	.1237	.6709	.9867	.6875	.0590
111.2498	.1662	.1235	.6714	.9893	.6866	.0541
120.1982	.1662	.1233	.6719	.9912	.6860	.0503
129.8306	.1663	.1231	.6724	.9928	.6855	.0467
142.0020	.1662	.1229	.6730	.9944	.6850	.0428
153.3004	.1662	.1228	.6735	.9956	.6846	.0398
167.5786	.1662	.1227	.6741	.9967	.6843	.0365
180.8339	.1661	.1226	.6746	.9974	.6840	.0339
200.0996	.1661	.1225	.6753	.9982	.6838	.0307

MACH NO = 10.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0680	.9095	1.1854	-.0792	1.0251	1.0125
1.0348	.0721	.8638	.9770	.0117	.9963	.9823
1.2863	.0764	.8092	.8164	.1116	.9646	.9453
1.6684	.0807	.7361	.6844	.2354	.9254	.8942
2.1282	.0835	.6612	.6018	.3558	.8873	.8395
2.6657	.0847	.5879	.5496	.4703	.8510	.7835
3.2758	.0847	.5193	.5184	.5746	.8180	.7284
3.9496	.0841	.4576	.5021	.6645	.7895	.6758
4.6751	.0834	.4037	.4968	.7377	.7663	.6271
5.4399	.0829	.3576	.4934	.7936	.7486	.5829
6.2327	.0828	.3187	.5076	.8334	.7360	.5431
7.0441	.0832	.2860	.5191	.8599	.7276	.5077
7.8673	.0841	.2587	.5326	.8755	.7227	.4762
9.5312	.0870	.2164	.5614	.8843	.7199	.4231
11.0330	.0908	.1888	.5864	.8770	.7222	.3844
12.5285	.0954	.1681	.6087	.8636	.7264	.3523
14.1786	.1012	.1507	.6297	.8469	.7317	.3226
15.6589	.1067	.1387	.6454	.8327	.7362	.2999
17.1500	.1125	.1292	.6583	.8207	.7400	.2801
18.8483	.1190	.1206	.6699	.8104	.7433	.2605
20.4481	.1249	.1142	.6780	.8044	.7452	.2443
22.1538	.1307	.1088	.6842	.8019	.7460	.2292
24.2259	.1370	.1035	.6888	.8035	.7455	.2132
26.3052	.1423	.0993	.6912	.8091	.7437	.1992
28.6476	.1479	.0956	.6920	.8184	.7408	.1855
31.6311	.1522	.0919	.6913	.8325	.7363	.1705
34.7536	.1561	.0890	.6897	.8480	.7314	.1573
38.3609	.1593	.0864	.6873	.8655	.7258	.1443
42.7982	.1621	.0839	.6844	.8848	.7197	.1310
47.1634	.1640	.0822	.6819	.9012	.7145	.1201
51.9108	.1655	.0807	.6796	.9162	.7098	.1102
57.6734	.1666	.0794	.6775	.9311	.7050	.1001
63.3391	.1673	.0785	.6759	.9429	.7013	.0919
69.5014	.1678	.0777	.6747	.9531	.6981	.0843
76.9860	.1681	.0770	.6737	.9628	.6950	.0766
84.3501	.1683	.0765	.6730	.9702	.6927	.0704
92.3654	.1684	.0760	.6726	.9764	.6907	.0646
102.1074	.1684	.0757	.6725	.9821	.6889	.0587
111.6991	.1683	.0754	.6725	.9863	.6876	.0539
122.1448	.1683	.0752	.6727	.9897	.6865	.0495
134.8475	.1682	.0750	.6730	.9926	.6856	.0450
147.3591	.1681	.0748	.6734	.9947	.6849	.0413
160.9886	.1680	.0747	.6739	.9962	.6844	.0379
177.5657	.1679	.0746	.6744	.9975	.6840	.0345
201.6124	.1678	.0745	.6752	.9986	.6837	.0305

MACH NO = 15.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0678	.9025	1.1854	-.0792	1.0251	1.0125
1.0874	.0727	.8451	.9351	.0344	.9891	.9744
1.3443	.0765	.7907	.7891	.1329	.9579	.9372
1.7312	.0801	.7185	.6675	.2551	.9192	.8863
2.1929	.0822	.6450	.5901	.3741	.8815	.8324
2.7278	.0826	.5736	.5405	.4872	.8457	.7775
3.4565	.0818	.4948	.5058	.6094	.8070	.7135
4.1237	.0806	.4369	.4915	.6957	.7796	.6635
4.8314	.0794	.3867	.4868	.7656	.7575	.6175
5.5668	.0785	.3439	.4892	.8191	.7405	.5761
6.3189	.0779	.3079	.4965	.8576	.7284	.5391
7.0790	.0777	.2776	.5069	.8835	.7201	.5063
7.9929	.0781	.2476	.5218	.9014	.7145	.4717
9.6495	.0802	.2063	.5515	.9083	.7123	.4197
11.2552	.0835	.1774	.5797	.8970	.7159	.3793
12.7924	.0878	.1567	.6045	.8783	.7218	.3472
14.2590	.0927	.1415	.6257	.8575	.7284	.3213
15.7932	.0985	.1291	.6451	.8353	.7354	.2980
17.1637	.1042	.1203	.6599	.8171	.7412	.2799
18.5286	.1101	.1132	.6721	.8018	.7460	.2639
19.9189	.1161	.1073	.6820	.7900	.7498	.2494
21.3687	.1221	.1023	.6897	.7822	.7522	.2359
22.9176	.1280	.0979	.6953	.7786	.7534	.2230
24.7686	.1343	.0936	.6993	.7796	.7530	.2093
26.6396	.1396	.0900	.7010	.7851	.7513	.1971
28.7398	.1446	.0867	.7011	.7946	.7483	.1850
31.1489	.1492	.0837	.6999	.8077	.7442	.1728
33.8908	.1533	.0808	.6976	.8235	.7391	.1607
37.3487	.1571	.0780	.6944	.8430	.7330	.1477
41.1212	.1602	.0757	.6910	.8623	.7269	.1357
45.6712	.1628	.0735	.6875	.8824	.7205	.1236
50.8101	.1649	.0717	.6843	.9011	.7146	.1123
56.4426	.1664	.0702	.6815	.9177	.7093	.1021
62.6211	.1675	.0691	.6793	.9323	.7047	.0928
70.0515	.1684	.0681	.6773	.9461	.7003	.0837
77.5626	.1688	.0674	.6758	.9568	.6969	.0761
85.8131	.1691	.0668	.6748	.9659	.6940	.0692
94.8784	.1693	.0663	.6741	.9734	.6917	.0630
104.8420	.1693	.0660	.6738	.9795	.6897	.0573
116.8439	.1693	.0657	.6736	.9849	.6880	.0517
128.9924	.1693	.0654	.6737	.9887	.6868	.0470
142.3518	.1692	.0652	.6740	.9916	.6859	.0427
157.0428	.1692	.0651	.6744	.9939	.6852	.0389
173.1976	.1691	.0650	.6748	.9955	.6846	.0354
201.3845	.1691	.0648	.6757	.9972	.6841	.0305

MACH NO = 20.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.0

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0677	.8999	1.1854	-.0792	1.0251	1.0125
1.0853	.0725	.8430	.3366	.0335	.9894	.9747
1.3398	.0761	.7891	.7907	.1315	.9583	.9378
1.7229	.0796	.7174	.6686	.2533	.9198	.8873
2.1797	.0815	.6445	.5905	.3721	.8821	.8338
2.7081	.0819	.5736	.5401	.4854	.8463	.7794
3.4270	.0809	.4953	.5043	.6081	.8074	.7159
4.0837	.0795	.4376	.4889	.6954	.7797	.6663
4.7789	.0781	.3876	.4831	.7667	.7571	.6207
5.4997	.0770	.3450	.4845	.8217	.7397	.5797
6.2340	.0762	.3091	.4907	.8620	.7269	.5431
6.9747	.0758	.2789	.5003	.8897	.7182	.5105
7.8612	.0758	.2490	.5143	.9095	.7119	.4764
9.6005	.0774	.2046	.5454	.9190	.7089	.4211
11.2603	.0805	.1745	.5751	.9070	.7127	.3791
12.9453	.0849	.1520	.6030	.8848	.7197	.3443
14.3977	.0897	.1372	.6248	.8616	.7271	.3190
15.7697	.0949	.1262	.6434	.8388	.7343	.2984
17.0881	.1005	.1176	.6591	.8180	.7409	.2809
18.4903	.1068	.1103	.6731	.7989	.7469	.2644
19.7936	.1127	.1047	.6836	.7851	.7513	.2507
21.1394	.1187	.0999	.6918	.7753	.7544	.2380
22.6899	.1250	.0954	.6983	.7695	.7562	.2248
24.2385	.1307	.0917	.7023	.7689	.7564	.2131
25.9187	.1361	.0882	.7043	.7726	.7553	.2016
27.9545	.1415	.0848	.7048	.7813	.7525	.1893
30.0907	.1461	.0819	.7038	.7931	.7488	.1779
32.4450	.1502	.0792	.7017	.8074	.7443	.1669
35.3346	.1541	.0765	.6988	.8249	.7387	.1550
38.4545	.1573	.0741	.6956	.8428	.7330	.1440
42.1874	.1602	.0718	.6921	.8618	.7270	.1327
46.6017	.1628	.0698	.6886	.8813	.7208	.1212
52.3011	.1651	.0679	.6852	.9010	.7146	.1094
58.3149	.1667	.0664	.6823	.9180	.7092	.0991
64.9638	.1679	.0653	.6800	.9329	.7045	.0897
72.9394	.1687	.0643	.6779	.9469	.7000	.0806
81.0913	.1692	.0636	.6764	.9578	.6966	.0730
90.0846	.1695	.0631	.6754	.9669	.6937	.0661
100.8819	.1697	.0626	.6747	.9749	.6912	.0594
111.9302	.1698	.0623	.6743	.9808	.6893	.0538
124.1399	.1698	.0620	.6743	.9855	.6878	.0487
137.6051	.1699	.0618	.6744	.9890	.6867	.0442
153.7935	.1698	.0616	.6748	.9919	.6858	.0397
170.3620	.1698	.0614	.6752	.9938	.6852	.0359
200.1964	.1698	.0613	.6760	.9959	.6845	.0307

MACH NO = 25.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 5.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0677	.8989	1.1854	-.0792	1.0251	1.0125
1.0843	.0724	.8422	.9373	.0331	.9895	.9748
1.3377	.0760	.7885	.7914	.1309	.9585	.9381
1.7191	.0794	.7171	.6691	.2525	.9200	.8878
2.2731	.0815	.6300	.5787	.3944	.8751	.8237
2.8121	.0815	.5601	.5322	.5061	.8397	.7695
3.4132	.0805	.4956	.5035	.6075	.8076	.7170
4.0652	.0790	.4381	.4877	.6952	.7798	.6676
4.8956	.0773	.3791	.4810	.7796	.7530	.6137
5.6125	.0761	.3380	.4830	.8324	.7363	.5737
6.3407	.0752	.3033	.4896	.8708	.7242	.5381
7.0720	.0748	.2742	.4991	.8971	.7158	.5065
7.9451	.0748	.2453	.5130	.9157	.7099	.4734
9.7870	.0765	.1995	.5461	.9238	.7074	.4159
11.5219	.0797	.1693	.5772	.9093	.7120	.3733
13.1295	.0839	.1486	.6038	.8865	.7192	.3409
14.6167	.0888	.1339	.6265	.8613	.7272	.3155
16.0089	.0942	.1231	.6457	.8366	.7350	.2950
17.3391	.1000	.1148	.6618	.8143	.7420	.2778
18.6413	.1060	.1082	.6751	.7955	.7480	.2627
19.9497	.1120	.1027	.6858	.7808	.7527	.2491
21.3019	.1181	.0980	.6941	.7705	.7559	.2365
22.7403	.1242	.0939	.7003	.7649	.7577	.2244
24.2839	.1300	.0902	.7042	.7640	.7580	.2128
25.9613	.1354	.0868	.7062	.7679	.7568	.2014
27.8358	.1405	.0836	.7065	.7762	.7541	.1900
29.9323	.1452	.0807	.7054	.7881	.7504	.1787
32.2188	.1492	.0780	.7033	.8023	.7459	.1679
34.7900	.1529	.0755	.7006	.8184	.7408	.1571
37.7771	.1562	.0731	.6973	.8362	.7351	.1463
41.3566	.1592	.0708	.6938	.8552	.7291	.1351
45.6129	.1619	.0687	.6902	.8746	.7230	.1238
50.3972	.1641	.0669	.6871	.8924	.7173	.1132
55.9457	.1660	.0654	.6842	.9094	.7119	.1029
62.5989	.1675	.0640	.6815	.9258	.7067	.0929
70.1255	.1685	.0630	.6792	.9404	.7021	.0836
78.4739	.1692	.0622	.6774	.9529	.6982	.0753
88.4898	.1696	.0615	.6760	.9639	.6947	.0673
98.8543	.1699	.0611	.6752	.9723	.6920	.0606
110.3647	.1700	.0607	.6748	.9789	.6899	.0545
123.1511	.1701	.0604	.6746	.9841	.6883	.0491
137.3558	.1701	.0601	.6748	.9879	.6871	.0442
153.1352	.1702	.0599	.6751	.9908	.6862	.0398
170.6622	.1702	.0598	.6756	.9929	.6855	.0359
201.4882	.1703	.0596	.6763	.9951	.6848	.0305

MACH NO = 30.00 CONE ANGLE = 0.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8436	.0677	.8982	1.1854	-.0792	1.0251	1.0125
1.0838	.0723	.8417	.9377	.0329	.9896	.9749
1.3365	.0759	.7881	.7918	.1305	.9587	.9383
1.7169	.0793	.7168	.6694	.2520	.9202	.8881
2.2694	.0813	.6299	.5787	.3939	.8752	.8241
2.8066	.0813	.5601	.5321	.5056	.8398	.7701
3.4056	.0803	.4957	.5032	.6071	.8077	.7176
4.0549	.0788	.4383	.4870	.6951	.7798	.6683
4.8813	.0769	.3794	.4800	.7799	.7529	.6146
5.5943	.0757	.3383	.4817	.8331	.7361	.5747
6.3180	.0748	.3037	.4880	.8721	.7238	.5392
7.0442	.0743	.2746	.4973	.8988	.7153	.5077
7.9104	.0742	.2457	.5110	.9180	.7092	.4746
9.7342	.0757	.1999	.5437	.9270	.7064	.4174
11.5731	.0790	.1678	.5767	.9117	.7112	.3722
13.1442	.0830	.1477	.6029	.8889	.7184	.3406
14.6990	.0881	.1324	.6269	.8617	.7270	.3143
16.0428	.0934	.1221	.6457	.8370	.7349	.2945
17.4199	.0994	.1136	.6627	.8131	.7425	.2768
18.6675	.1052	.1073	.6757	.7943	.7484	.2624
20.0151	.1116	.1017	.6869	.7785	.7534	.2485
21.3071	.1175	.0972	.6950	.7683	.7566	.2365
22.7874	.1238	.0930	.7014	.7622	.7586	.2241
24.2565	.1294	.0895	.7052	.7613	.7589	.2130
25.9768	.1350	.0860	.7073	.7653	.7576	.2013
27.7633	.1399	.0830	.7075	.7733	.7550	.1904
29.8903	.1447	.0800	.7063	.7856	.7511	.1789
32.0334	.1486	.0774	.7042	.7992	.7469	.1687
34.6233	.1523	.0749	.7014	.8156	.7416	.1578
37.4114	.1555	.0726	.6983	.8325	.7363	.1475
40.7259	.1584	.0704	.6948	.8507	.7305	.1369
44.9362	.1613	.0682	.6912	.8704	.7243	.1255
49.2732	.1635	.0664	.6882	.8873	.7189	.1155
54.6222	.1655	.0648	.6853	.9043	.7135	.1052
60.5557	.1670	.0635	.6827	.9198	.7086	.0957
68.2790	.1683	.0623	.6801	.9359	.7035	.0857
76.3128	.1691	.0615	.6782	.9489	.6994	.0773
85.9368	.1696	.0608	.6766	.9605	.6957	.0691
95.8806	.1699	.0603	.6757	.9694	.6929	.0623
107.8050	.1701	.0599	.6751	.9769	.6905	.0558
120.1368	.1702	.0595	.6748	.9824	.6888	.0503
134.9305	.1703	.0593	.6749	.9867	.6874	.0450
150.2295	.1704	.0590	.6753	.9896	.6865	.0405
168.5795	.1705	.0588	.6757	.9920	.6858	.0363
200.2440	.1706	.0586	.6765	.9944	.6850	.0307

MACH NO = 3.50 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0696	.9974	1.2101	-.0882	1.0311	1.0154
1.0127	.0757	.9470	1.0008	-.0004	1.0001	.9826
1.1814	.0807	.9056	.8838	.0655	.9769	.9547
1.5256	.0894	.8309	.7478	.1736	.9388	.9024
1.8634	.0961	.7684	.6777	.2572	.9093	.8564
2.2609	.1020	.7055	.6309	.3371	.8811	.8079
2.8447	.1079	.6295	.5942	.4306	.8482	.7459
3.3823	.1119	.5730	.5783	.4968	.8248	.6966
3.9826	.1156	.5215	.5713	.5539	.8047	.6488
4.8201	.1198	.4648	.5706	.6126	.7839	.5921
5.5586	.1230	.4258	.5742	.6505	.7706	.5497
6.3569	.1260	.3922	.5800	.6812	.7598	.5102
7.4373	.1298	.3569	.5889	.7109	.7493	.4650
8.8528	.1341	.3230	.6001	.7375	.7399	.4167
10.3965	.1381	.2968	.6106	.7575	.7329	.3742
11.7809	.1411	.2795	.6183	.7709	.7281	.3429
13.5617	.1444	.2631	.6264	.7842	.7234	.3096
15.4773	.1473	.2503	.6332	.7956	.7194	.2803
17.5354	.1498	.2403	.6389	.8055	.7159	.2544
19.7460	.1520	.2325	.6437	.8142	.7129	.2314
22.1225	.1539	.2262	.6477	.8220	.7101	.2110
24.2413	.1552	.2220	.6507	.8280	.7080	.1956
26.9666	.1566	.2179	.6538	.8345	.7057	.1788
29.9122	.1578	.2145	.6565	.8404	.7036	.1636
33.1020	.1589	.2118	.6589	.8458	.7017	.1498
36.5612	.1597	.2096	.6609	.8507	.7000	.1373
40.3168	.1605	.2078	.6628	.8552	.6984	.1258
44.3971	.1611	.2063	.6645	.8592	.6970	.1154
48.0672	.1615	.2053	.6658	.8622	.6959	.1074
52.8236	.1619	.2043	.6673	.8655	.6948	.0985
57.9974	.1623	.2035	.6686	.8684	.6938	.0904
63.6266	.1626	.2028	.6699	.8710	.6928	.0829
69.7524	.1628	.2022	.6710	.8733	.6920	.0761
76.4194	.1630	.2017	.6721	.8753	.6913	.0699
83.6762	.1631	.2014	.6731	.8771	.6907	.0641
90.2119	.1632	.2011	.6739	.8784	.6902	.0597
98.6902	.1633	.2008	.6748	.8797	.6898	.0548
107.9203	.1634	.2006	.6756	.8809	.6893	.0503
117.9692	.1634	.2004	.6764	.8819	.6890	.0462
128.9101	.1634	.2003	.6772	.8828	.6887	.0424
140.8228	.1634	.2002	.6779	.8836	.6884	.0390
151.5547	.1634	.2001	.6784	.8841	.6882	.0363
165.4801	.1634	.2000	.6791	.8846	.6880	.0333
180.6439	.1634	.1999	.6797	.8850	.6879	.0306
200.0484	.1634	.1998	.6803	.8854	.6877	.0277

MACH NO = 5.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0685	.9505	1.2101	-.0882	1.0311	1.0154
1.0206	.0737	.8987	.9928	.0036	.9987	.9813
1.2065	.0779	.8537	.8661	.0768	.9729	.9507
1.5834	.0845	.7735	.7231	.1960	.9309	.8942
1.9408	.0889	.7087	.6524	.2856	.8993	.8465
2.4598	.0929	.6300	.5970	.3894	.8627	.7857
2.9327	.0952	.5707	.5708	.4641	.8363	.7373
3.5926	.0976	.5033	.5542	.5438	.8082	.6791
4.1719	.0993	.4554	.5502	.5959	.7899	.6350
4.9547	.1015	.4033	.5530	.6466	.7720	.5839
5.6242	.1035	.3676	.5595	.6766	.7614	.5462
6.3293	.1058	.3367	.5683	.6987	.7536	.5115
7.2580	.1090	.3039	.5809	.7178	.7469	.4719
8.6448	.1140	.2673	.5990	.7334	.7414	.4231
10.1360	.1195	.2391	.6157	.7417	.7385	.3808
11.7422	.1252	.2172	.6300	.7467	.7367	.3437
13.2222	.1300	.2023	.6401	.7505	.7353	.3154
15.0860	.1354	.1885	.6495	.7556	.7335	.2858
17.1217	.1405	.1775	.6566	.7620	.7313	.2592
19.3371	.1451	.1689	.6618	.7696	.7286	.2354
21.6897	.1490	.1623	.6654	.7781	.7256	.2144
24.2025	.1523	.1571	.6679	.7871	.7224	.1958
26.9022	.1551	.1529	.6695	.7964	.7191	.1791
29.8146	.1574	.1496	.6704	.8059	.7158	.1640
32.4998	.1590	.1473	.6710	.8139	.7130	.1522
35.8760	.1606	.1451	.6713	.8228	.7098	.1396
39.5406	.1618	.1433	.6716	.8313	.7068	.1280
43.5222	.1627	.1418	.6717	.8391	.7041	.1175
47.8510	.1634	.1406	.6719	.8462	.7016	.1078
52.5590	.1640	.1396	.6721	.8526	.6993	.0989
57.6805	.1644	.1388	.6724	.8582	.6974	.0908
63.2524	.1647	.1381	6727	.8631	.6956	.0834
69.3150	.1649	.1376	.6731	.8674	.6941	.0766
74.9351	.1650	.1372	.6735	.8706	.6930	.0712
82.0278	.1651	.1368	.6740	.8738	.6919	.0653
89.7465	.1651	.1365	.6745	.8765	.6909	.0600
98.1469	.1651	.1362	.6751	.8788	.6901	.0551
107.2900	.1650	.1360	.6757	.8806	.6894	.0506
117.2419	.1650	.1358	.6763	.8822	.6889	.0465
128.0749	.1650	.1357	.6769	.8834	.6885	.0427
139.8675	.1649	.1356	.6775	.8843	.6881	.0392
153.8039	.1649	.1355	.6781	.8850	.6879	.0365
164.6113	.1648	.1354	.6787	.8856	.6877	.0335
179.6433	.1648	.1353	.6793	.8860	.6876	.0308
200.9459	.1647	.1352	.6800	.8863	.6874	.0276

MACH NO = 10.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0676	.9140	1.2101	-.0882	1.0311	1.0154
1.0080	.0716	.8660	1.0027	-.0013	1.0005	.9834
1.2415	.0755	.8101	.8439	.0916	.9677	.9452
1.5921	.0796	.7365	.7119	.2048	.9278	.8930
2.0092	.0823	.6620	.6284	.3128	.8897	.8380
2.4918	.0836	.5899	.5753	.4139	.8540	.7822
3.0349	.0838	.5229	.5428	.5048	.8220	.7277
3.6301	.0835	.4629	.5253	.5825	.7946	.6761
4.2664	.0831	.4106	.5186	.6453	.7724	.6284
4.9327	.0830	.3659	.5199	.6930	.7556	.5852
5.6192	.0832	.3281	.5265	.7270	.7436	.5465
6.3178	.0838	.2964	.5365	.7496	.7357	.5120
7.0229	.0848	.2698	.5485	.7631	.7309	.4814
8.4378	.0881	.2286	.5747	.7711	.7281	.4298
9.8443	.0925	.1990	.6000	.7646	.7304	.3884
11.2325	.0978	.1773	.6222	.7524	.7347	.3546
12.6013	.1036	.1611	.6409	.7391	.7393	.3267
13.9603	.1097	.1487	.6563	.7269	.7437	.3030
15.3291	.1159	.1389	.6687	.7168	.7472	.2823
16.7351	.1223	.1310	.6787	.7095	.7498	.2639
18.2115	.1285	.1244	.6863	.7053	.7513	.2469
19.7969	.1347	.1189	.6917	.7048	.7514	.2310
21.5364	.1405	.1140	.6951	.7081	.7503	.2157
23.4834	.1459	.1098	.6967	.7152	.7478	.2008
25.6932	.1509	.1060	.6966	.7259	.7440	.1862
28.2367	.1552	.1026	.6952	.7396	.7392	.1719
31.2094	.1589	.0995	.6928	.7559	.7334	.1577
34.6996	.1619	.0969	.6897	.7738	.7271	.1437
38.6853	.1642	.0946	.6865	.7915	.7209	.1305
43.0607	.1658	.0928	.6837	.8078	.7151	.1186
47.8606	.1669	.0914	.6812	.8222	.7100	.1078
53.1264	.1676	.0903	.6792	.8348	.7056	.0980
58.9052	.1681	.0894	.6777	.8456	.7018	.0891
65.2493	.1683	.0887	.6766	.8547	.6986	.0810
72.2160	.1683	.0882	.6759	.8622	.6959	.0737
79.8688	.1683	.0877	.6755	.8684	.6938	.0670
88.2771	.1682	.0874	.6754	.8733	.6920	.0610
97.5175	.1681	.0871	.6756	.8772	.6907	.0555
107.6737	.1679	.0869	.6759	.8802	.6896	.0504
118.8380	.1678	.0867	.6763	.8824	.6888	.0459
131.1111	.1677	.0865	.6769	.8841	.6882	.0417
144.6038	.1675	.0864	.6775	.8853	.6878	.0380
159.4371	.1674	.0863	.6782	.8861	.6875	.0345
175.7442	.1674	.0863	.6789	.8866	.6873	.0314
201.3302	.1673	.0862	.6798	.8870	.6872	.0275

MACH NO = 15.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0674	.9070	1.2101	-.0882	1.0311	1.0154
1.0569	.0721	.8469	.9615	.0198	.9930	.9752
1.2947	.0756	.7914	.8168	.1111	.9608	.9369
1.6490	.0791	.7188	.6951	.2225	.9215	.8851
2.0671	.0811	.6459	.6169	.3290	.8840	.8309
2.5466	.0817	.5758	.5665	.4286	.8489	.7763
3.1943	.0813	.4988	.5305	.5347	.8114	.7131
3.7824	.0804	.4425	.5152	.6088	.7853	.6640
4.4024	.0795	.3937	.5094	.6685	.7642	.6191
5.0428	.0789	.3522	.5107	.7139	.7482	.5786
5.6944	.0787	.3171	.5168	.7465	.7368	.5425
6.3499	.0788	.2877	.5260	.7683	.7291	.5105
7.1347	.0795	.2584	.5395	.7833	.7238	.4768
8.5495	.0820	.2179	.5669	.7891	.7217	.4261
10.0348	.0862	.1873	.5953	.7785	.7255	.3834
11.4458	.0912	.1659	.6198	.7615	.7315	.3500
12.7858	.0968	.1504	.6404	.7431	.7380	.3232
14.0728	.1029	.1388	.6576	.7256	.7441	.3012
15.3335	.1092	.1298	.6717	.7104	.7495	.2823
16.4916	.1151	.1231	.6823	.6992	.7534	.2669
17.7878	.1215	.1171	.6913	.6906	.7565	.2515
19.1528	.1279	.1119	.6980	.6860	.7581	.2372
20.6281	.1341	.1074	.7025	.6856	.7582	.2234
22.2453	.1399	.1033	.7049	.6894	.7569	.2100
23.8849	.1449	.1000	.7054	.6965	.7544	.1980
25.9101	.1498	.0966	.7045	.7078	.7504	.1849
28.2133	.1541	.0935	.7023	.7222	.7453	.1720
30.8365	.1579	.0906	.6993	.7386	.7395	.1593
33.9155	.1610	.0880	.6958	.7565	.7332	.1466
37.6283	.1637	.0856	.6920	.7754	.7266	.1338
41.6998	.1658	.0837	.6885	.7927	.7204	.1221
46.8156	.1675	.0820	.6852	.8103	.7142	.1100
52.4963	.1686	.0806	.6824	.8257	.7088	.0991
58.7859	.1692	.0796	.6802	.8389	.7042	.0892
65.7546	.1696	.0788	.6785	.8500	.7002	.0804
72.8039	.1697	.0783	.6775	.8584	.6973	.0731
81.2943	.1697	.0778	.6767	.8659	.6946	.0659
90.7105	.1696	.0774	.6764	.8719	.6925	.0594
101.1559	.1695	.0772	.6764	.8764	.6909	.0535
112.7449	.1694	.0769	.6766	.8798	.6897	.0483
125.8037	.1692	.0768	.6771	.8822	.6889	.0435
138.6249	.1691	.0766	.6776	.8838	.6883	.0396
154.3190	.1690	.0765	.6783	.8850	.6879	.0357
171.7316	.1689	.0764	.6790	.8858	.6876	.0321
201.4375	.1688	.0763	.6800	.8865	.6874	.0275

MACH NO = 20.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID CA	AERODYNAMIC XCP/L	COEFFICIENTS YCP/D	XVCP/LV	RN/RB
.8264	.0673	.9044	1.2101	-.0882	1.0311	1.0154
1.0549	.0719	.8448	.9630	.0190	.9933	.9755
1.2905	.0753	.7898	.9184	.1098	.9613	.9375
1.6414	.0786	.7178	.8963	.2208	.9221	.8861
2.1451	.0807	.6311	.6053	.3477	.8774	.8215
2.6305	.0810	.5623	.5584	.4458	.8428	.7675
3.1679	.0804	.4992	.5293	.5335	.8119	.7155
3.7472	.0794	.4431	.5130	.6084	.7854	.6667
4.4809	.0782	.3857	.5058	.6796	.7603	.6138
5.1114	.0774	.3456	.5072	.7236	.7448	.5746
5.7496	.0770	.3119	.5132	.7553	.7337	.5397
6.3888	.0770	.2834	.5222	.7765	.7262	.5087
7.1503	.0775	.2552	.5353	.7912	.7210	.4762
8.7547	.0801	.2103	.5667	.7960	.7193	.4197
10.2688	.0842	.1804	.5960	.7828	.7239	.3774
11.6807	.0892	.1598	.6210	.7636	.7307	.3450
12.9999	.0948	.1451	.6420	.7431	.7379	.3194
14.1566	.1004	.1349	.6583	.7251	.7443	.2998
15.3720	.1066	.1263	.6730	.7080	.7503	.2817
16.5797	.1131	.1195	.6849	.6943	.7552	.2658
17.8089	.1195	.1138	.6941	.6844	.7586	.2513
19.0982	.1259	.1089	.7010	.6787	.7606	.2377
20.4851	.1321	.1046	.7056	.6774	.7611	.2247
21.9866	.1379	.1008	.7082	.6803	.7601	.2120
23.6474	.1433	.0973	.7088	.6874	.7576	.1996
25.5210	.1482	.0940	.7078	.6983	.7537	.1873
27.5847	.1524	.0910	.7057	.7118	.7490	.1753
29.9052	.1561	.0883	.7028	.7271	.7436	.1636
32.5982	.1594	.0858	.6993	.7440	.7376	.1518
35.8173	.1622	.0834	.6956	.7620	.7313	.1398
39.5561	.1647	.0812	.6919	.7798	.7250	.1280
43.8274	.1666	.0794	.6886	.7965	.7191	.1167
48.8854	.1682	.0779	.6856	.8125	.7135	.1057
54.4785	.1692	.0767	.6830	.8268	.7084	.0957
61.2615	.1699	.0757	.6807	.8402	.7037	.0859
68.8108	.1703	.0750	.6790	.8515	.6997	.0771
77.2157	.1704	.0744	.6778	.8606	.6965	.0692
86.5760	.1703	.0740	.6771	.8679	.6939	.0621
97.0040	.1703	.0737	.6768	.8735	.6920	.0557
109.6240	.1701	.0734	.6769	.8776	.6905	.0500
121.5731	.1700	.0732	.6773	.8806	.6895	.0449
136.0034	.1699	.0730	.6778	.8826	.6887	.0403
152.0835	.1699	.0729	.6785	.8841	.6882	.0362
170.0013	.1698	.0728	.6792	.8850	.6879	.0325
201.6444	.1697	.0727	.6802	.8859	.6876	.0275

MACH NO = 25.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0673	.9034	1.2101	-.0882	1.0311	1.0154
1.0540	.0718	.8441	.9637	.0187	.9934	.9757
1.2886	.0752	.7893	.8192	.1092	.9615	.9378
1.6378	.0784	.7175	.6969	.2201	.9224	.8866
2.1390	.0805	.6311	.6055	.3469	.8777	.8222
2.6216	.0807	.5624	.5582	.4450	.8431	.7684
3.1557	.0800	.4994	.5287	.5329	.8121	.7166
3.7308	.0789	.4435	.5119	.6082	.7855	.6680
4.4585	.0776	.3862	.5042	.6801	.7602	.6153
5.0831	.0767	.3462	.5051	.7248	.7444	.5762
5.7145	.0763	.3124	.5107	.7570	.7330	.5415
6.3461	.0761	.2840	.5193	.7790	.7253	.5107
7.0975	.0765	.2558	.5321	.7943	.7199	.4783
8.6754	.0788	.2109	.5629	.8003	.7178	.4222
10.2667	.0829	.1791	.5941	.7865	.7226	.3774
11.7328	.0880	.1577	.6204	.7657	.7300	.3439
12.9939	.0933	.1437	.6410	.7450	.7373	.3195
14.2724	.0995	.1326	.6596	.7238	.7447	.2980
15.5032	.1060	.1241	.6749	.7054	.7512	.2799
16.6342	.1122	.1177	.6863	.6917	.7561	.2651
17.8731	.1188	.1121	.6959	.6811	.7598	.2506
19.0765	.1249	.1075	.7025	.6752	.7619	.2379
20.4657	.1313	.1032	.7073	.6734	.7625	.2248
21.9663	.1372	.0994	.7098	.6763	.7615	.2122
23.5046	.1423	.0961	.7104	.6829	.7592	.2007
25.3543	.1473	.0928	.7094	.6938	.7553	.1883
27.3713	.1517	.0899	.7072	.7073	.7506	.1765
29.4620	.1552	.0873	.7045	.7215	.7456	.1657
32.0572	.1585	.0847	.7010	.7383	.7396	.1540
35.1567	.1615	.0823	.6971	.7564	.7333	.1421
38.4307	.1639	.0802	.6937	.7727	.7275	.1313
42.3799	.1660	.0784	.6904	.7892	.7217	.1203
46.6299	.1676	.0769	.6876	.8038	.7165	.1104
52.1474	.1690	.0755	.6847	.8193	.7111	.0997
58.9512	.1700	.0744	.6820	.8343	.7058	.0890
66.0840	.1704	.0736	.6800	.8463	.7015	.0801
74.6521	.1707	.0729	.6785	.8570	.6978	.0714
84.2504	.1707	.0724	.6776	.8654	.6948	.0637
94.1976	.1706	.0721	.6772	.8714	.6927	.0573
106.1584	.1705	.0718	.6772	.8761	.6910	.0511
119.5677	.1705	.0715	.6775	.8795	.6898	.0456
133.4690	.1704	.0714	.6780	.8817	.6891	.0410
150.1844	.1704	.0712	.6786	.8833	.6885	.0365
167.5117	.1703	.0711	.6793	.8844	.6881	.0329
201.3618	.1702	.0709	.6803	.8855	.6877	.0275

MACH NO = 30.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.8264	.0673	.9027	1.2101	-.0882	1.0311	1.0154
1.0535	.0718	.8435	.9641	.0185	.9935	.9757
1.2875	.0751	.7889	.8196	.1089	.9615	.9380
1.7130	.0788	.7027	.6785	.2413	.9149	.8763
2.1356	.0803	.6310	.6056	.3464	.8778	.8226
2.6168	.0805	.5624	.5581	.4446	.8432	.7689
3.2607	.0796	.4878	.5240	.5488	.8065	.7072
3.8402	.0784	.4334	.5092	.6216	.7808	.6595
4.4462	.0773	.3864	.5033	.6803	.7601	.6161
5.1927	.0762	.3392	.5048	.7328	.7416	.5699
5.8208	.0758	.3067	.5108	.7632	.7308	.5361
6.4476	.0757	.2792	.5197	.7837	.7236	.5061
7.1916	.0760	.2519	.5325	.7977	.7187	.4746
8.8643	.0786	.2058	.5655	.8016	.7173	.4163
10.4185	.0827	.1758	.5959	.7866	.7226	.3737
11.8445	.0876	.1556	.6216	.7655	.7300	.3416
13.1587	.0933	.1413	.6431	.7432	.7379	.3165
14.3922	.0993	.1309	.6611	.7221	.7453	.2961
15.5792	.1057	.1228	.6760	.7038	.7518	.2789
16.7528	.1122	.1163	.6879	.6893	.7569	.2636
17.9466	.1186	.1110	.6971	.6789	.7606	.2498
19.1979	.1250	.1063	.7039	.6728	.7627	.2367
20.5377	.1312	.1022	.7084	.6712	.7633	.2242
21.9800	.1369	.0986	.7108	.6741	.7623	.2121
23.5754	.1423	.0952	.7113	.6812	.7598	.2002
25.3424	.1471	.0920	.7102	.6919	.7560	.1884
27.2526	.1512	.0892	.7080	.7048	.7514	.1772
29.3821	.1548	.0866	.7052	.7195	.7463	.1661
31.8334	.1581	.0841	.7018	.7357	.7406	.1550
34.7388	.1610	.0817	.6981	.7530	.7345	.1436
38.0099	.1635	.0796	.6946	.7697	.7286	.1326
41.6372	.1656	.0778	.6914	.7852	.7231	.1222
45.8020	.1674	.0763	.6886	.8000	.7179	.1122
50.7762	.1688	.0749	.6858	.8146	.7127	.1021
57.4517	.1699	.0737	.6829	.8303	.7072	.0912
64.8926	.1705	.0728	.6807	.8437	.7025	.0814
73.2142	.1708	.0721	.6790	.8547	.6986	.0727
82.5238	.1709	.0716	.6779	.8636	.6955	.0650
92.9430	.1708	.0712	.6774	.8703	.6931	.0580
104.6079	.1708	.0709	.6773	.8752	.6914	.0519
117.6684	.1707	.0706	.6776	.8786	.6902	.0463
132.2911	.1707	.0704	.6781	.8811	.6893	.0414
148.6620	.1707	.0703	.6787	.8828	.6887	.0370
166.9895	.1706	.0701	.6794	.8840	.6882	.0330
200.3131	.1706	.0700	.6804	.8852	.6878	.0277

MACH NO = 3.50 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0666	1.0274	1.3492	-.1340	1.0718	1.0353
.9026	.0724	.9637	1.1232	-.0551	1.0295	.9909
1.0993	.0786	.8968	.9630	.0192	.9897	.9417
1.3216	.0846	.8321	.8571	.0842	.9549	.8917
1.6350	.0912	.7562	.7725	.1543	.9173	.8296
1.9143	.0959	.7002	.7289	.2027	.8914	.7811
2.2988	.1008	.6369	.6933	.2548	.8634	.7229
2.6349	.1042	.5916	.6754	.2903	.8444	.6787
3.0875	.1083	.5421	.6628	.3264	.8251	.6271
3.4740	.1114	.5079	.6579	.3499	.8125	.5889
3.9856	.1151	.4712	.6562	.3733	.8000	.5449
4.4165	.1181	.4462	.6573	.3880	.7921	.5126
4.9814	.1217	.4198	.6603	.4025	.7843	.4757
5.8223	.1267	.3900	.6659	.4179	.7760	.4297
6.8615	.1321	.3642	.6725	.4311	.7690	.3837
7.8516	.1365	.3468	.6777	.4406	.7639	.3483
9.0948	.1410	.3315	.6827	.4504	.7587	.3121
10.3008	.1445	.3212	.6860	.4585	.7543	.2835
11.6413	.1476	.3129	.6885	.4665	.7500	.2573
13.3733	.1505	.3056	.6905	.4756	.7451	.2298
15.0526	.1524	.3007	.6917	.4833	.7410	.2083
17.1571	.1542	.2965	.6927	.4914	.7366	.1864
19.1812	.1553	.2937	.6933	.4979	.7332	.1693
21.3961	.1561	.2916	.6939	.5038	.7300	.1538
24.1881	.1567	.2897	.6945	.5097	.7269	.1380
26.8844	.1570	.2884	.6951	.5142	.7245	.1255
30.2884	.1572	.2873	.6958	.5185	.7221	.1126
33.5793	.1573	.2865	.6966	.5217	.7204	.1024
37.1926	.1573	.2860	.6973	.5244	.7190	.0932
41.7583	.1573	.2854	.6983	.5268	.7177	.0836
46.1751	.1572	.2851	.6992	.5285	.7168	.0761
51.7574	.1572	.2848	.7003	.5299	.7160	.0683
57.1584	.1571	.2845	.7013	.5309	.7155	.0622
63.0917	.1570	.2844	.7022	.5316	.7151	.0566
70.5920	.1569	.2842	.7033	.5321	.7148	.0508
77.8497	.1569	.2841	.7042	.5325	.7147	.0462
85.8233	.1568	.2840	.7050	.5327	.7145	.0421
95.9038	.1568	.2840	.7059	.5328	.7145	.0378
105.6588	.1568	.2839	.7067	.5329	.7144	.0344
117.9916	.1567	.2839	.7075	.5329	.7144	.0309
129.9266	.1567	.2838	.7081	.5329	.7144	.0281
143.0395	.1567	.2838	.7087	.5329	.7144	.0256
159.6179	.1567	.2838	.7093	.5329	.7144	.0230
175.6615	.1567	.2838	.7098	.5329	.7144	.0209
201.3661	.1567	.2837	.7104	.5328	.7145	.0183

MACH NO = 5.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0660	.9801	1.3492	-.1340	1.0718	1.0353
.9309	.0716	.9065	1.0926	-.0424	1.0227	.9835
1.1117	.0759	.8457	.9510	.0256	.9863	.9388
1.3688	.0808	.7715	.8323	.1012	.9458	.8818
1.6598	.0849	.7012	.7558	.1672	.9104	.8250
1.9829	.0883	.6365	.7067	.2239	.8800	.7700
2.4095	.0916	.5673	.6710	.2805	.8497	.7077
2.7939	.0941	.5171	.6550	.3179	.8296	.6596
3.1993	.0966	.4737	.6475	.3471	.8140	.6155
3.6211	.0992	.4367	.6459	.3690	.8023	.5755
4.0558	.1021	.4054	.6482	.3847	.7938	.5393
4.5009	.1052	.3789	.6529	.3958	.7879	.5068
5.0471	.1091	.3525	.6600	.4047	.7831	.4718
5.8912	.1152	.3215	.6715	.4124	.7790	.4263
6.7694	.1216	.2982	.6822	.4167	.7767	.3874
7.6896	.1279	.2804	.6910	.4201	.7749	.3536
8.6641	.1339	.2667	.6977	.4240	.7728	.3237
9.7112	.1394	.2559	.7023	.4290	.7701	.2968
10.7223	.1439	.2482	.7050	.4345	.7672	.2747
11.9771	.1484	.2412	.7065	.4421	.7631	.2515
13.3907	.1522	.2355	.7066	.4510	.7593	.2296
15.0046	.1552	.2308	.7057	.4611	.7529	.2089
16.8683	.1575	.2271	.7042	.4719	.7471	.1891
19.0372	.1591	.2241	.7022	.4830	.7412	.1704
21.4844	.1601	.2218	.7004	.4933	.7356	.1533
24.1950	.1606	.2201	.6989	.5024	.7308	.1379
27.1991	.1607	.2188	.6979	.5100	.7267	.1241
30.5298	.1607	.2178	.6974	.5163	.7233	.1118
34.2235	.1606	.2170	.6973	.5212	.7207	.1006
38.3204	.1604	.2165	.6976	.5250	.7186	.0906
42.8653	.1601	.2160	.6982	.5279	.7171	.0816
47.9078	.1599	.2157	.6990	.5300	.7160	.0735
53.5031	.1597	.2154	.7000	.5314	.7152	.0662
59.7124	.1595	.2152	.7010	.5323	.7147	.0596
66.6037	.1594	.2150	.7021	.5329	.7144	.0537
74.2523	.1593	.2149	.7032	.5333	.7142	.0484
82.7419	.1592	.2148	.7042	.5334	.7141	.0436
92.1652	.1591	.2147	.7052	.5335	.7141	.0393
102.6251	.1591	.2147	.7061	.5335	.7141	.0354
114.2357	.1590	.2146	.7070	.5334	.7142	.0319
127.1237	.1590	.2146	.7078	.5333	.7142	.0287
141.4298	.1590	.2146	.7085	.5332	.7142	.0259
157.3098	.1590	.2145	.7091	.5331	.7143	.0233
174.9369	.1589	.2145	.7097	.5331	.7143	.0210
201.4925	.1589	.2145	.7103	.5330	.7144	.0183

MACH NO = 10.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0655	.9433	1.3492	-.1340	1.0718	1.0353
.9118	.0696	.8770	1.1110	-.0500	1.0268	.9885
1.1169	.0735	.8076	.9439	.0294	.9843	.9376
1.4025	.0771	.7256	.8129	.1148	.9385	.8748
1.7284	.0797	.6484	.7307	.1892	.8986	.8127
2.0888	.0815	.5785	.6797	.2521	.8649	.7536
2.4771	.0826	.5171	.6491	.3037	.8372	.6988
2.8842	.0837	.4646	.6329	.3437	.8158	.6493
3.3011	.0850	.4206	.6269	.3729	.8002	.6054
3.7200	.0866	.3841	.6276	.3927	.7896	.5668
4.1391	.0886	.3539	.6327	.4051	.7829	.5329
4.4846	.0905	.3327	.6389	.4112	.7796	.5079
4.8944	.0932	.3111	.6477	.4148	.7777	.4811
5.7613	.0999	.2756	.6679	.4140	.7782	.4327
6.6001	.1073	.2508	.6858	.4089	.7809	.3944
7.4183	.1148	.2329	.7003	.4035	.7838	.3630
8.2304	.1224	.2195	.7117	.3991	.7861	.3364
8.9915	.1294	.2099	.7197	.3967	.7874	.3148
9.8487	.1366	.2015	.7257	.3965	.7875	.2936
10.7628	.1433	.1947	.7292	.3991	.7861	.2739
11.7604	.1494	.1890	.7303	.4047	.7831	.2552
12.7829	.1543	.1845	.7295	.4124	.7790	.2385
14.0312	.1587	.1803	.7269	.4233	.7732	.2209
15.4666	.1621	.1766	.7228	.4364	.7661	.2036
17.1255	.1645	.1734	.7179	.4510	.7583	.1867
19.0787	.1661	.1708	.7127	.4662	.7502	.1701
21.1702	.1668	.1688	.7080	.4800	.7428	.1553
23.8485	.1669	.1670	.7035	.4939	.7353	.1397
27.0582	.1665	.1657	.7000	.5062	.7287	.1247
30.7957	.1659	.1648	.6977	.5161	.7234	.1109
34.8066	.1653	.1642	.6966	.5231	.7197	.0991
38.7720	.1647	.1637	.6964	.5275	.7173	.0896
43.4908	.1642	.1634	.6969	.5306	.7156	.0805
48.8297	.1638	.1631	.6979	.5325	.7146	.0722
55.0363	.1635	.1628	.6992	.5336	.7140	.0645
62.4367	.1633	.1626	.7006	.5341	.7138	.0572
70.2688	.1631	.1625	.7020	.5343	.7137	.0510
79.8164	.1630	.1623	.7034	.5342	.7137	.0451
90.6099	.1629	.1622	.7047	.5341	.7138	.0399
102.8121	.1628	.1621	.7059	.5339	.7139	.0353
115.4850	.1627	.1621	.7069	.5338	.7140	.0315
130.9346	.1627	.1620	.7078	.5336	.7141	.0279
148.4011	.1627	.1620	.7086	.5334	.7141	.0247
168.1478	.1626	.1620	.7094	.5333	.7142	.0218
201.7369	.1626	.1619	.7103	.5331	.7143	.0182

MACH NO = 15.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0654	.9363	1.3492	-.1340	1.0718	1.0353
.9456	.0700	.8579	1.0765	-.0355	1.0190	.9797
1.1517	.0734	.7898	.9223	.0416	.9777	.9294
1.4368	.0766	.7097	.7999	.1247	.9332	.8678
1.7598	.0787	.6345	.7221	.1974	.8942	.8072
2.1142	.0800	.5666	.6734	.2589	.8613	.7497
2.5577	.0808	.4981	.6402	.3167	.8303	.6884
2.9527	.0815	.4488	.6261	.3542	.8102	.6416
3.3527	.0824	.4075	.6211	.3813	.7957	.6004
3.7515	.0837	.3733	.6223	.3997	.7858	.5642
4.1456	.0854	.3448	.6275	.4111	.7797	.5324
4.5328	.0874	.3211	.6352	.4172	.7764	.5046
4.9741	.0902	.2982	.6458	.4195	.7752	.4762
5.9372	.0978	.2602	.6703	.4150	.7776	.4241
6.7866	.1055	.2364	.6898	.4071	.7818	.3867
7.6506	.1140	.2186	.7064	.3987	.7863	.3549
8.4969	.1227	.2057	.7191	.3920	.7899	.3285
9.3540	.1311	.1959	.7282	.3883	.7919	.3055
10.2503	.1391	.1882	.7339	.3881	.7920	.2846
11.2169	.1465	.1819	.7365	.3917	.7901	.2650
12.2155	.1526	.1768	.7364	.3985	.7864	.2475
13.4085	.1580	.1721	.7339	.4091	.7807	.2294
14.7539	.1622	.1681	.7296	.4225	.7736	.2118
16.2881	.1653	.1646	.7242	.4377	.7655	.1949
18.0582	.1673	.1616	.7183	.4536	.7569	.1784
20.0551	.1685	.1592	.7127	.4691	.7486	.1628
22.3882	.1689	.1574	.7075	.4839	.7407	.1478
24.9285	.1688	.1560	.7034	.4965	.7339	.1343
27.9849	.1683	.1550	.7000	.5079	.7278	.1210
31.4559	.1675	.1543	.6976	.5172	.7228	.1087
35.0903	.1668	.1538	.6964	.5238	.7193	.0983
38.8740	.1661	.1535	.6961	.5282	.7169	.0894
42.0040	.1656	.1532	.6964	.5311	.7154	.0815
47.0000	.1652	.1530	.6971	.5327	.7145	.0748
51.8259	.1650	.1527	.6982	.5337	.7140	.0682
57.2741	.1648	.1525	.6994	.5342	.7137	.0621
63.5458	.1646	.1523	.7008	.5343	.7137	.0562
70.9202	.1645	.1522	.7021	.5343	.7137	.0506
79.8017	.1644	.1521	.7034	.5342	.7137	.0451
90.7869	.1643	.1519	.7047	.5341	.7138	.0399
103.7160	.1642	.1519	.7059	.5340	.7138	.0350
119.9598	.1641	.1518	.7071	.5338	.7139	.0304
138.6795	.1640	.1517	.7081	.5336	.7140	.0264
160.2524	.1640	.1517	.7091	.5333	.7142	.0229
200.7280	.1640	.1516	.7104	.5329	.7144	.0183

MACH NO = 20.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0653	.9336	1.3492	-.1340	1.0718	1.0353
.9439	.0699	.8559	1.0781	-.0361	1.0194	.9802
1.1481	.0732	.7883	.9241	.0406	.9783	.9303
1.4305	.0762	.7087	.8013	.1235	.9338	.8691
1.7502	.0782	.6338	.7229	.1962	.8949	.8089
2.1007	.0793	.5663	.6734	.2578	.8618	.7518
2.5385	.0801	.4980	.6395	.3101	.8336	.6908
2.9278	.0806	.4488	.6247	.3540	.8103	.6444
3.3212	.0814	.4076	.6191	.3816	.7955	.6034
3.7128	.0825	.3733	.6199	.4005	.7854	.5675
4.0989	.0840	.3448	.6247	.4123	.7790	.5360
4.4773	.0858	.3211	.6321	.4187	.7756	.5084
4.9074	.0885	.2982	.6425	.4212	.7743	.4802
5.8966	.0961	.2582	.6686	.4163	.7769	.4260
6.8654	.1049	.2310	.6915	.4062	.7823	.3836
7.7762	.1142	.2129	.7095	.3962	.7877	.3508
8.6645	.1236	.2000	.7230	.3885	.7918	.3237
9.5661	.1327	.1903	.7322	.3846	.7939	.3003
10.4616	.1407	.1831	.7372	.3850	.7937	.2801
11.4932	.1485	.1768	.7392	.3896	.7912	.2599
12.6406	.1551	.1714	.7380	.3986	.7864	.2407
13.9339	.1604	.1668	.7343	.4113	.7796	.2222
15.3886	.1643	.1629	.7290	.4264	.7715	.2045
16.9699	.1669	.1596	.7231	.4420	.7631	.1882
18.8318	.1686	.1568	.7170	.4583	.7544	.1720
20.9569	.1695	.1547	.7114	.4739	.7461	.1567
23.4567	.1697	.1530	.7063	.4884	.7382	.1418
26.0824	.1694	.1519	.7024	.5004	.7318	.1289
29.2334	.1688	.1510	.6992	.5112	.7260	.1163
32.7115	.1679	.1505	.6971	.5197	.7215	.1049
36.2989	.1671	.1501	.6961	.5256	.7183	.0953
40.0192	.1665	.1497	.6959	.5295	.7162	.0870
43.7385	.1660	.1495	.6963	.5318	.7150	.0801
47.9868	.1657	.1493	.6972	.5333	.7142	.0734
52.6571	.1654	.1491	.6983	.5340	.7138	.0672
57.8697	.1653	.1489	.6995	.5342	.7137	.0614
63.7739	.1652	.1487	.7009	.5342	.7137	.0560
70.1652	.1651	.1485	.7021	.5341	.7138	.0511
78.0670	.1650	.1484	.7033	.5340	.7138	.0461
87.5497	.1649	.1483	.7045	.5340	.7138	.0413
99.2539	.1648	.1482	.7056	.5340	.7139	.0365
114.1609	.1647	.1481	.7067	.5339	.7139	.0319
132.2821	.1647	.1480	.7078	.5336	.7140	.0276
155.0337	.1647	.1480	.7090	.5331	.7143	.0236
200.0179	.1659	.1475	.7125	.5290	.7165	.0184

MACH NO = 25.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0653	.9326	1.3492	-.1340	1.0718	1.0353
.9389	.0697	.8566	1.0830	-.0382	1.0205	.9815
1.1767	.0735	.7785	.9077	.0501	.9731	.9237
1.4521	.0763	.7021	.7941	.1293	.9307	.8647
1.8159	.0783	.6190	.7108	.2094	.8878	.7975
2.1568	.0792	.5555	.6671	.2669	.8570	.7434
2.5769	.0798	.4915	.6371	.3207	.8281	.6860
2.9465	.0804	.4455	.6239	.3559	.8093	.6423
3.3788	.0812	.4010	.6185	.3853	.7935	.5978
3.7449	.0822	.3695	.6196	.4022	.7845	.5647
4.1627	.0839	.3393	.6253	.4143	.7780	.5311
4.5112	.0856	.3178	.6324	.4196	.7751	.5060
4.9051	.0880	.2970	.6421	.4217	.7740	.4804
6.4879	.1009	.2391	.6830	.4104	.7801	.3991
7.8626	.1148	.2100	.7115	.3947	.7885	.3479
9.1451	.1284	.1930	.7293	.3847	.7938	.3108
10.4798	.1410	.1815	.7383	.3835	.7945	.2797
11.8602	.1512	.1735	.7401	.3906	.7907	.2535
13.3751	.1588	.1672	.7370	.4042	.7834	.2298
15.0446	.1641	.1622	.7310	.4216	.7741	.2084
16.8743	.1675	.1582	.7242	.4400	.7642	.1891
18.8701	.1694	.1552	.7177	.4574	.7549	.1717
21.1175	.1704	.1529	.7118	.4736	.7462	.1556
23.4963	.1707	.1514	.7070	.4873	.7388	.1416
26.0980	.1705	.1503	.7031	.4992	.7325	.1289
28.9484	.1699	.1496	.7000	.5094	.7270	.1173
32.0742	.1691	.1491	.6978	.5177	.7226	.1068
35.5031	.1683	.1487	.6964	.5242	.7191	.0973
39.2650	.1675	.1483	.6959	.5289	.7166	.0886
43.5322	.1669	.1480	.6961	.5321	.7148	.0804
48.0747	.1664	.1478	.6968	.5340	.7138	.0733
53.0591	.1660	.1476	.6978	.5350	.7133	.0667
58.5282	.1658	.1474	.6991	.5353	.7131	.0608
64.5294	.1656	.1472	.7004	.5354	.7131	.0554
71.1149	.1654	.1471	.7016	.5353	.7131	.0504
78.5864	.1652	.1469	.7028	.5352	.7132	.0458
86.5418	.1651	.1469	.7038	.5350	.7133	.0417
95.2732	.1650	.1468	.7047	.5349	.7133	.0380
104.8560	.1649	.1467	.7056	.5348	.7134	.0346
115.3733	.1647	.1467	.7063	.5347	.7134	.0316
126.9159	.1646	.1467	.7070	.5346	.7135	.0288
139.5835	.1645	.1466	.7077	.5345	.7136	.0262
153.9564	.1644	.1466	.7083	.5343	.7137	.0238
169.2601	.1644	.1466	.7089	.5342	.7137	.0217
200.1761	.1643	.1466	.7098	.5339	.7139	.0184

MACH NO = 30.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7412	.0653	.9319	1.3492	-.1340	1.0718	1.0353
.9384	.0697	.8561	1.0833	-.0384	1.0206	.9816
1.1758	.0734	.7781	.9082	.0499	.9733	.9239
1.4505	.0762	.7018	.7944	.1290	.9309	.8651
1.8134	.0781	.6189	.7110	.2091	.8879	.7979
2.1533	.0790	.5554	.6670	.2667	.8571	.7439
2.5722	.0796	.4915	.6369	.3206	.8282	.6866
2.9405	.0801	.4454	.6235	.3559	.8093	.6430
3.3712	.0809	.4009	.6179	.3855	.7934	.5986
3.7357	.0819	.3695	.6190	.4025	.7843	.5655
4.1516	.0835	.3393	.6245	.4146	.7778	.5320
4.4982	.0852	.3178	.6316	.4201	.7749	.5069
4.8898	.0876	.2969	.6412	.4222	.7737	.4813
6.5070	.1007	.2377	.6834	.4103	.7801	.3983
7.8607	.1145	.2091	.7117	.3943	.7887	.3480
9.1617	.1284	.1919	.7300	.3839	.7943	.3103
10.4728	.1410	.1807	.7390	.3825	.7950	.2798
11.8737	.1513	.1725	.7408	.3897	.7911	.2532
13.3644	.1589	.1663	.7376	.4033	.7839	.2300
15.0605	.1643	.1612	.7314	.4211	.7743	.2082
16.9213	.1677	.1572	.7244	.4399	.7642	.1886
18.8861	.1696	.1542	.7179	.4572	.7550	.1716
21.0943	.1706	.1520	.7120	.4731	.7465	.1558
23.5065	.1709	.1504	.7072	.4871	.7390	.1415
26.0619	.1707	.1494	.7033	.4988	.7327	.1290
28.9517	.1701	.1486	.7001	.5092	.7271	.1173
32.1252	.1693	.1481	.6978	.5178	.7225	.1067
35.4971	.1685	.1477	.6964	.5242	.7191	.0973
39.3155	.1677	.1474	.6959	.5290	.7165	.0885
43.5106	.1670	.1471	.6960	.5322	.7148	.0805
48.1197	.1665	.1469	.6967	.5341	.7138	.0732
53.0184	.1662	.1466	.6978	.5350	.7133	.0668
58.5663	.1659	.1464	.6991	.5353	.7131	.0608
64.6621	.1657	.1463	.7004	.5353	.7131	.0553
71.1413	.1656	.1461	.7016	.5352	.7132	.0504
78.4804	.1654	.1460	.7028	.5351	.7132	.0459
86.5458	.1653	.1459	.7038	.5350	.7133	.0417
95.1200	.1651	.1459	.7047	.5349	.7133	.0381
104.8329	.1650	.1458	.7056	.5348	.7134	.0347
115.5073	.1649	.1458	.7063	.5347	.7134	.0315
127.2380	.1648	.1457	.7071	.5346	.7135	.0287
139.7079	.1647	.1457	.7077	.5345	.7136	.0262
153.8334	.1646	.1457	.7083	.5343	.7137	.0238
169.3568	.1645	.1457	.7089	.5342	.7137	.0217
200.3112	.1644	.1456	.7098	.5339	.7139	.0184

MACH NO = 3.50 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RR
.6580	.0634	1.0680	1.5198	-.1820	1.1325	1.0642
.8300	.0704	.9809	1.2299	-.0952	1.0693	.9977
.9903	.0762	.9139	1.0778	-.0363	1.0264	.9428
1.2015	.0827	.8409	.9594	.0214	.9844	.8791
1.4330	.0884	.7757	.8832	.0685	.9501	.8185
1.6835	.0935	.7184	.8330	.1071	.9221	.7616
1.9506	.0980	.6689	.7999	.1384	.8993	.7091
2.2327	.1021	.6263	.7779	.1639	.8807	.6610
2.5308	.1058	.5896	.7634	.1846	.8656	.6168
2.8413	.1096	.5587	.7551	.2006	.8540	.5766
3.1633	.1132	.5327	.7503	.2133	.8447	.5401
3.4968	.1168	.5109	.7482	.2232	.8375	.5069
3.9015	.1210	.4898	.7475	.2323	.8309	.4716
4.4508	.1261	.4681	.7479	.2418	.8240	.4310
5.0386	.1310	.4513	.7489	.2496	.8183	.3946
5.6736	.1354	.4381	.7496	.2569	.8130	.3616
6.3673	.1393	.4277	.7495	.2643	.8076	.3314
7.1342	.1427	.4195	.7488	.2718	.8021	.3033
7.9922	.1454	.4131	.7474	.2797	.7964	.2771
8.9636	.1475	.4080	.7456	.2877	.7906	.2524
10.0744	.1491	.4041	.7436	.2958	.7847	.2290
11.3559	.1501	.4010	.7415	.3037	.7789	.2069
12.8442	.1506	.3986	.7395	.3112	.7735	.1860
14.5806	.1508	.3968	.7380	.3179	.7685	.1665
16.6109	.1507	.3954	.7371	.3236	.7644	.1482
18.9456	.1505	.3944	.7369	.3282	.7611	.1317
21.5704	.1502	.3937	.7372	.3314	.7587	.1169
24.5218	.1500	.3932	.7380	.3337	.7571	.1039
27.8410	.1499	.3928	.7391	.3352	.7560	.0923
31.5744	.1496	.3925	.7404	.3362	.7553	.0820
35.7741	.1494	.3923	.7417	.3368	.7549	.0729
40.4989	.1493	.3921	.7430	.3370	.7547	.0648
45.8140	.1492	.3920	.7443	.3372	.7546	.0576
51.7962	.1491	.3920	.7455	.3372	.7546	.0511
58.5265	.1491	.3919	.7465	.3371	.7546	.0455
66.0996	.1491	.3919	.7475	.3370	.7547	.0404
74.6213	.1490	.3918	.7484	.3369	.7547	.0359
84.2105	.1490	.3918	.7492	.3368	.7548	.0319
95.0009	.1490	.3918	.7499	.3368	.7548	.0283
107.1430	.1490	.3918	.7505	.3367	.7549	.0252
120.8061	.1490	.3918	.7510	.3367	.7549	.0224
136.1807	.1490	.3917	.7515	.3366	.7549	.0199
153.4813	.1490	.3917	.7519	.3366	.7550	.0177
172.9490	.1490	.3917	.7523	.3366	.7550	.0157
200.0847	.1490	.3917	.7526	.3366	.7550	.0136

MACH NO = 5.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6590	.0632	1.0206	1.5198	-.1820	1.1325	1.0642
.8119	.0686	.9417	1.2512	-.1024	1.0746	1.0043
.9866	.0737	.8666	1.0762	-.0355	1.0258	.9440
1.2144	.0791	.7859	.9463	.0285	.9792	.8755
1.4655	.0837	.7139	.8650	.0802	.9416	.8106
1.7357	.0877	.6513	.8140	.1212	.9119	.7508
2.0195	.0913	.5982	.7829	.1528	.8888	.6967
2.3138	.0947	.5535	.7641	.1769	.8712	.6484
2.6159	.0983	.5161	.7545	.1944	.8585	.6052
2.9230	.1020	.4852	.7505	.2067	.8495	.5669
3.2339	.1060	.4597	.7500	.2153	.8433	.5327
3.5035	.1094	.4413	.7514	.2205	.8395	.5062
3.8220	.1136	.4231	.7540	.2248	.8363	.4782
4.3336	.1202	.4004	.7587	.2296	.8329	.4391
4.8639	.1266	.3831	.7625	.2338	.8298	.4048
5.4206	.1325	.3696	.7649	.2384	.8265	.3741
6.0138	.1379	.3591	.7658	.2438	.8225	.3461
6.6567	.1427	.3507	.7653	.2501	.8179	.3202
7.3654	.1468	.3440	.7635	.2576	.8124	.2958
8.1592	.1500	.3387	.7605	.2663	.8062	.2725
9.0619	.1524	.3344	.7565	.2759	.7991	.2501
10.1033	.1540	.3309	.7519	.2863	.7916	.2284
11.2010	.1548	.3284	.7477	.2958	.7846	.2093
12.6151	.1550	.3262	.7433	.3060	.7772	.1890
14.2958	.1547	.3245	.7396	.3153	.7705	.1694
16.3022	.1542	.3231	.7371	.3230	.7648	.1507
18.6990	.1536	.3221	.7359	.3290	.7605	.1332
21.4682	.1530	.3214	.7359	.3329	.7576	.1175
24.6038	.1526	.3208	.7368	.3353	.7559	.1036
28.1552	.1523	.3204	.7383	.3366	.7550	.0913
32.1788	.1520	.3201	.7399	.3372	.7546	.0806
36.7383	.1519	.3198	.7415	.3374	.7544	.0711
41.9060	.1518	.3197	.7431	.3374	.7544	.0627
47.2003	.1517	.3196	.7444	.3374	.7544	.0559
53.7653	.1517	.3195	.7457	.3372	.7545	.0493
61.2076	.1517	.3194	.7469	.3371	.7546	.0435
69.6447	.1516	.3193	.7479	.3370	.7547	.0384
79.2096	.1516	.3193	.7488	.3369	.7548	.0339
90.0530	.1516	.3193	.7496	.3368	.7548	.0299
102.3458	.1516	.3193	.7503	.3367	.7549	.0263
116.2819	.1516	.3193	.7508	.3367	.7549	.0232
132.0807	.1516	.3192	.7514	.3367	.7549	.0205
149.9912	.1516	.3192	.7518	.3366	.7550	.0181
170.2956	.1516	.3192	.7522	.3366	.7550	.0160
200.1093	.1516	.3192	.7526	.3366	.7550	.0136

MACH NO = 10.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0631	.9876	1.5198	-.1820	1.1325	1.0642
.8221	.0679	.8998	1.2368	-.0974	1.0709	1.0006
1.0116	.0723	.8189	1.0538	-.0255	1.0185	.9360
1.2554	.0764	.7334	.9221	.0423	.9692	.8642
1.5227	.0797	.6579	.8414	.0963	.9299	.7972
1.8052	.0824	.5935	.7928	.1378	.8997	.7368
2.0943	.0850	.5401	.7648	.1684	.8774	.6838
2.3852	.0877	.4963	.7501	.1900	.8617	.6376
2.6737	.0907	.4605	.7445	.2041	.8514	.5976
2.9577	.0941	.4313	.7446	.2126	.8452	.5628
3.2363	.0980	.4075	.7481	.2170	.8420	.5324
3.5093	.1021	.3880	.7534	.2189	.8407	.5057
3.7773	.1063	.3717	.7590	.2194	.8403	.4819
4.3028	.1149	.3467	.7691	.2192	.8404	.4412
4.8214	.1233	.3287	.7769	.2191	.8405	.4073
5.3457	.1314	.3153	.7822	.2200	.8399	.3779
5.8556	.1385	.3057	.7849	.2224	.8381	.3532
6.4344	.1452	.2975	.7851	.2272	.8346	.3287
7.0653	.1509	.2910	.7828	.2345	.8293	.3056
7.7214	.1551	.2859	.7787	.2434	.8228	.2849
8.5114	.1585	.2815	.7724	.2550	.8144	.2633
9.4189	.1604	.2778	.7648	.2683	.8047	.2422
10.3972	.1612	.2751	.7572	.2815	.7951	.2230
11.5878	.1610	.2728	.7494	.2953	.7851	.2033
12.9645	.1602	.2712	.7426	.3080	.7758	.1845
14.5230	.1590	.2699	.7374	.3185	.7682	.1671
16.1699	.1579	.2691	.7344	.3261	.7626	.1518
18.0927	.1569	.2683	.7330	.3315	.7587	.1373
20.1853	.1562	.2677	.7331	.3348	.7563	.1243
22.3389	.1558	.2672	.7340	.3365	.7551	.1132
24.9082	.1555	.2668	.7354	.3374	.7544	.1024
27.8542	.1553	.2664	.7372	.3377	.7542	.0923
31.0890	.1552	.2661	.7390	.3377	.7542	.0832
35.2632	.1551	.2658	.7408	.3376	.7542	.0739
40.5317	.1550	.2655	.7427	.3375	.7544	.0647
47.3534	.1550	.2653	.7445	.3373	.7545	.0558
54.9776	.1549	.2652	.7460	.3371	.7546	.0483
64.4146	.1549	.2651	.7474	.3369	.7547	.0414
75.4173	.1550	.2650	.7486	.3367	.7549	.0355
87.3849	.1550	.2650	.7496	.3366	.7550	.0308
102.1993	.1550	.2649	.7505	.3364	.7551	.0264
119.4722	.1551	.2649	.7512	.3364	.7551	.0226
138.2600	.1551	.2648	.7518	.3363	.7552	.0196
161.5168	.1552	.2648	.7522	.3364	.7552	.0168
201.8693	.1552	.2648	.7528	.3364	.7551	.0135

MACH NO = 15.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0631	.9765	1.5198	-.1820	1.1325	1.0642
.8389	.0682	.9849	1.2157	-.0900	1.0655	.9945
1.0481	.0725	.7975	1.0276	-.0134	1.0097	.9245
1.3136	.0764	.7079	.8986	.0555	.9589	.8487
1.6024	.0794	.6303	.8223	.1109	.9193	.7792
1.9035	.0819	.5656	.7784	.1514	.8898	.7179
2.1728	.0841	.5186	.7568	.1772	.8710	.6707
2.4710	.0868	.4758	.7452	.1968	.8567	.6252
2.7607	.0899	.4416	.7422	.2087	.8480	.5865
3.0400	.0935	.4143	.7444	.2151	.8434	.5535
3.3086	.0974	.3922	.7493	.2178	.8415	.5251
3.5391	.1010	.3761	.7546	.2184	.8410	.5029
3.7902	.1052	.3610	.7605	.2182	.8412	.4808
4.7289	.1214	.3212	.7792	.2156	.8430	.4130
5.6222	.1360	.2994	.7887	.2163	.8426	.3641
6.5454	.1479	.2859	.7897	.2233	.8375	.3244
7.5498	.1583	.2768	.7839	.2366	.8278	.2900
8.6645	.1612	.2704	.7740	.2541	.8151	.2595
9.9016	.1631	.2660	.7627	.2729	.8013	.2323
11.3080	.1632	.2630	.7520	.2912	.7880	.2076
12.8190	.1622	.2611	.7436	.3063	.7770	.1864
14.4870	.1608	.2599	.7377	.3181	.7684	.1674
16.3330	.1596	.2591	.7343	.3265	.7623	.1505
18.3800	.1587	.2585	.7331	.3319	.7584	.1353
20.6527	.1580	.2579	.7335	.3349	.7562	.1217
23.1771	.1576	.2574	.7348	.3364	.7551	.1095
25.9815	.1574	.2569	.7364	.3372	.7546	.0985
29.0973	.1572	.2566	.7380	.3376	.7542	.0886
32.5591	.1570	.2563	.7395	.3379	.7540	.0797
36.4052	.1567	.2561	.7408	.3382	.7538	.0717
40.6793	.1565	.2559	.7420	.3385	.7536	.0645
45.4259	.1563	.2558	.7431	.3386	.7535	.0580
50.7012	.1561	.2558	.7441	.3387	.7535	.0522
56.5632	.1559	.2557	.7451	.3386	.7535	.0470
63.0774	.1557	.2557	.7461	.3385	.7536	.0423
70.3165	.1556	.2556	.7470	.3383	.7537	.0380
78.3611	.1555	.2556	.7478	.3382	.7538	.0342
87.3006	.1554	.2556	.7485	.3380	.7540	.0308
97.2343	.1553	.2555	.7492	.3378	.7541	.0277
108.2731	.1553	.2555	.7498	.3377	.7542	.0249
120.5400	.1552	.2555	.7504	.3375	.7543	.0224
134.1716	.1552	.2555	.7509	.3374	.7544	.0202
149.3200	.1551	.2555	.7513	.3373	.7545	.0182
166.1539	.1551	.2555	.7517	.3372	.7546	.0163
200.2455	.1551	.2555	.7523	.3370	.7547	.0136

MACH NO = 20.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0631	.9738	1.5198	-.1820	1.1325	1.0642
.8375	.0680	.8828	1.2173	-.0905	1.0659	.9950
1.0451	.0723	.7959	1.0292	-.0141	1.0103	.9254
1.3084	.0761	.7066	.8998	.0557	.9594	.8500
1.5947	.0789	.6292	.8229	.1103	.9197	.7809
1.8929	.0813	.5646	.7785	.1509	.8901	.7199
2.1923	.0837	.5123	.7546	.1796	.8693	.6675
2.4860	.0863	.4705	.7438	.1984	.8556	.6230
2.7705	.0894	.4372	.7413	.2097	.8473	.5853
3.0441	.0929	.4104	.7439	.2156	.8431	.5530
3.3067	.0967	.3888	.7491	.2179	.8414	.5253
3.5315	.1003	.3730	.7545	.2183	.8411	.5036
3.7760	.1044	.3581	.7605	.2179	.8414	.4820
4.7333	.1212	.3172	.7805	.2144	.8439	.4127
5.6142	.1361	.2957	.7904	.2144	.8439	.3645
6.5488	.1485	.2821	.7915	.2214	.8388	.3243
7.5689	.1572	.2729	.7852	.2352	.8288	.2894
8.6733	.1620	.2666	.7749	.2531	.8158	.2593
9.9319	.1638	.2621	.7630	.2728	.8014	.2317
11.2836	.1637	.2592	.7523	.2908	.7883	.2080
12.8128	.1626	.2574	.7434	.3065	.7769	.1864
14.4553	.1612	.2563	.7374	.3184	.7682	.1677
16.3220	.1598	.2555	.7339	.3270	.7619	.1506
18.3375	.1589	.2548	.7328	.3323	.7581	.1356
20.6357	.1582	.2542	.7333	.3352	.7560	.1218
23.1210	.1579	.2537	.7347	.3365	.7550	.1097
25.9564	.1577	.2532	.7364	.3371	.7546	.0985
29.1106	.1575	.2529	.7381	.3375	.7544	.0885
32.5225	.1573	.2526	.7396	.3378	.7541	.0798
36.4154	.1571	.2524	.7409	.3381	.7539	.0717
40.6261	.1568	.2523	.7420	.3384	.7537	.0646
45.4305	.1566	.2522	.7431	.3386	.7535	.0580
50.6279	.1564	.2521	.7441	.3386	.7535	.0523
56.5591	.1562	.2520	.7451	.3386	.7535	.0470
62.9762	.1561	.2520	.7461	.3385	.7536	.0423
70.2996	.1559	.2519	.7470	.3383	.7537	.0380
78.2229	.1558	.2519	.7478	.3381	.7539	.0343
87.2649	.1557	.2519	.7486	.3380	.7540	.0308
97.3260	.1557	.2519	.7492	.3378	.7541	.0277
108.2109	.1556	.2519	.7498	.3376	.7542	.0249
120.6329	.1556	.2519	.7504	.3375	.7543	.0224
134.0722	.1555	.2518	.7509	.3374	.7544	.0202
149.4097	.1555	.2518	.7513	.3373	.7545	.0182
166.0036	.1555	.2518	.7517	.3371	.7546	.0164
200.2486	.1554	.2518	.7523	.3370	.7547	.0136

MACH NO = 25.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID CA	AERODYNAMIC XCP/L	COEFFICIENTS YCP/D	XVCP/LV	RN/RB
.6580	.0631	.9728	1.5198	-.1820	1.1325	1.0642
.8368	.0680	.8821	1.2180	-.0908	1.0661	.9952
1.0437	.0722	.7954	1.0300	-.0145	1.0106	.9258
1.3061	.0759	.7062	.9003	.0554	.9597	.8506
1.5912	.0787	.6288	.8232	.1100	.9200	.7816
1.8880	.0810	.5643	.7785	.1507	.8903	.7208
2.1858	.0834	.5120	.7544	.1795	.8693	.6685
2.4778	.0859	.4701	.7434	.1984	.8556	.6242
2.7604	.0889	.4367	.7409	.2098	.8473	.5865
3.0319	.0924	.4099	.7434	.2157	.8430	.5544
3.2922	.0962	.3883	.7486	.2180	.8413	.5267
3.5421	.1002	.3706	.7547	.2183	.8411	.5027
3.7831	.1042	.3560	.7608	.2178	.8415	.4814
4.7479	.1214	.3150	.7813	.2138	.8444	.4118
5.6345	.1365	.2936	.7914	.2136	.8445	.3635
6.5511	.1488	.2804	.7923	.2205	.8395	.3242
7.5789	.1576	.2711	.7859	.2346	.8292	.2891
8.6942	.1624	.2648	.7752	.2529	.8159	.2588
9.9304	.1642	.2604	.7632	.2725	.8017	.2318
11.3299	.1640	.2574	.7520	.2913	.7880	.2073
12.8319	.1628	.2556	.7432	.3068	.7767	.1862
14.4875	.1613	.2545	.7371	.3189	.7679	.1674
16.3688	.1599	.2538	.7336	.3275	.7616	.1502
18.4031	.1589	.2531	.7326	.3326	.7579	.1352
20.6601	.1583	.2526	.7332	.3353	.7559	.1217
23.1651	.1580	.2520	.7347	.3366	.7550	.1095
26.0227	.1578	.2515	.7365	.3371	.7546	.0983
29.1177	.1577	.2512	.7382	.3374	.7544	.0885
32.5535	.1575	.2509	.7396	.3377	.7542	.0797
36.4734	.1572	.2507	.7409	.3381	.7539	.0716
40.7190	.1570	.2505	.7420	.3384	.7537	.0644
45.4322	.1568	.2504	.7431	.3386	.7535	.0580
50.8102	.1565	.2504	.7442	.3386	.7535	.0521
56.6362	.1564	.2503	.7451	.3386	.7535	.0469
63.1052	.1562	.2503	.7461	.3385	.7536	.0423
70.2883	.1561	.2502	.7470	.3383	.7537	.0380
78.4853	.1560	.2502	.7478	.3381	.7539	.0342
87.3654	.1559	.2502	.7486	.3380	.7540	.0308
97.2251	.1558	.2502	.7492	.3378	.7541	.0277
108.4764	.1558	.2501	.7499	.3376	.7542	.0249
120.6653	.1557	.2501	.7504	.3375	.7543	.0224
134.1993	.1557	.2501	.7509	.3374	.7544	.0202
149.6440	.1556	.2501	.7513	.3372	.7545	.0181
166.3762	.1556	.2501	.7517	.3371	.7546	.0163
200.5016	.1556	.2501	.7523	.3370	.7547	.0136

MACH NO = 30.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 5.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.0631	.9721	1.5198	-.1820	1.1325	1.0642
.8365	.0680	.9816	1.2184	-.0909	1.0662	.9954
1.0430	.0722	.7950	1.0304	-.0147	1.0107	.9261
1.3047	.0758	.7059	.9006	.0552	.9598	.8510
1.5892	.0786	.6285	.8233	.1098	.9201	.7821
1.8852	.0809	.5640	.7785	.1506	.8904	.7213
2.1822	.0832	.5117	.7543	.1794	.8694	.6691
2.4732	.0857	.4699	.7432	.1984	.8556	.6248
2.7547	.0887	.4365	.7406	.2098	.8473	.5872
3.0251	.0921	.4096	.7431	.2157	.8430	.5552
3.2841	.0959	.3880	.7483	.2180	.8413	.5276
3.5327	.0998	.3703	.7545	.2183	.8411	.5035
3.7722	.1039	.3557	.7607	.2178	.8415	.4824
4.7296	.1210	.3146	.7813	.2135	.8446	.4129
5.6296	.1364	.2928	.7919	.2131	.8449	.3637
6.5601	.1490	.2793	.7928	.2201	.8398	.3238
7.5501	.1576	.2704	.7865	.2337	.8299	.2900
8.6809	.1625	.2639	.7756	.2524	.8163	.2591
9.9362	.1643	.2594	.7633	.2724	.8017	.2317
11.3206	.1641	.2565	.7521	.2911	.7881	.2074
12.8043	.1629	.2547	.7433	.3066	.7768	.1865
14.4829	.1614	.2536	.7370	.3190	.7678	.1675
16.3417	.1600	.2529	.7335	.3275	.7616	.1504
18.3493	.1590	.2522	.7325	.3326	.7579	.1355
20.6359	.1584	.2516	.7331	.3354	.7558	.1218
23.1776	.1581	.2511	.7347	.3366	.7550	.1095
26.0032	.1579	.2506	.7365	.3370	.7547	.0984
29.0599	.1578	.2502	.7382	.3373	.7544	.0887
32.5432	.1576	.2499	.7397	.3377	.7542	.0797
36.4161	.1573	.2497	.7409	.3380	.7539	.0717
40.7220	.1571	.2496	.7421	.3384	.7537	.0644
45.3804	.1568	.2495	.7431	.3386	.7535	.0581
50.6896	.1566	.2494	.7441	.3386	.7535	.0522
56.5937	.1564	.2494	.7451	.3386	.7535	.0470
62.9824	.1563	.2493	.7461	.3385	.7536	.0423
70.2646	.1562	.2493	.7470	.3383	.7537	.0381
78.3630	.1561	.2493	.7478	.3381	.7539	.0342
87.3689	.1560	.2492	.7486	.3380	.7540	.0308
97.1137	.1559	.2492	.7492	.3378	.7541	.0277
108.2208	.1559	.2492	.7498	.3376	.7542	.0249
120.5727	.1558	.2492	.7504	.3375	.7543	.0224
134.3091	.1558	.2492	.7509	.3374	.7544	.0202
149.1729	.1557	.2492	.7513	.3373	.7545	.0182
166.1153	.1557	.2492	.7517	.3371	.7546	.0164
200.3241	.1557	.2492	.7523	.3370	.7547	.0136

MACH NO = 3.50 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0292	1.0542
.8894	.1420	.9729	1.1243	-.0556	1.0097	1.0062
1.1626	.1603	.9352	.8756	.0710	.9876	.9823
1.6052	.1867	.8797	.6919	.2340	.9591	.9463
2.3074	.2181	.8025	.5768	.4366	.9236	.8943
3.1105	.2413	.7277	.5227	.6246	.8907	.8414
4.0938	.2600	.6516	.4954	.8104	.8582	.7846
5.2544	.2748	.5789	.4852	.9829	.8280	.7267
6.7915	.2887	.5037	.4871	1.1531	.7982	.6620
8.3151	.2993	.4463	.4958	1.2751	.7769	.6083
10.0017	.3087	.3971	.5077	1.3743	.7595	.5582
11.8501	.3172	.3555	.5207	1.4539	.7456	.5120
14.1623	.3261	.3161	.5358	1.5251	.7331	.4640
16.0398	.3323	.2914	.5466	1.5676	.7257	.4311
18.0442	.3381	.2702	.5567	1.6031	.7195	.4008
20.1803	.3433	.2521	.5659	1.6332	.7142	.3729
22.4537	.3481	.2366	.5743	1.6591	.7097	.3471
24.4580	.3517	.2253	.5806	1.6781	.7064	.3272
27.0022	.3556	.2136	.5876	1.6983	.7028	.3050
29.7079	.3592	.2035	.5938	1.7162	.6997	.2845
32.5881	.3623	.1947	.5995	1.7323	.6969	.2654
35.6576	.3651	.1872	.6046	1.7468	.6943	.2478
38.9332	.3675	.1806	.6092	1.7600	.6920	.2313
42.4323	.3697	.1749	.6134	1.7720	.6899	.2160
46.1730	.3716	.1700	.6173	1.7828	.6880	.2018
50.1735	.3732	.1657	.6208	1.7927	.5863	.1885
54.4525	.3747	.1620	.6241	1.8016	.6848	.1760
59.0291	.3760	.1587	.6272	1.8096	.6834	.1645
63.9234	.3771	.1559	.6300	1.8167	.6821	.1536
69.1567	.3781	.1535	.6327	1.8232	.6810	.1435
74.7512	.3790	.1514	.6352	1.8288	.6800	.1341
80.7307	.3798	.1496	.6375	1.8338	.6791	.1253
87.1200	.3806	.1480	.6397	1.8382	.6784	.1171
92.7768	.3811	.1469	.6414	1.8414	.6778	.1107
99.9883	.3818	.1457	.6434	1.8447	.6772	.1035
107.6906	.3824	.1447	.6453	1.8476	.6767	.0967
115.9164	.3830	.1438	.6470	1.8502	.6763	.0904
124.7013	.3835	.1430	.6486	1.8524	.6759	.0846
134.0836	.3840	.1424	.6502	1.8545	.6755	.0791
144.1050	.3845	.1418	.6515	1.8565	.6752	.0739
154.8101	.3848	.1413	.6528	1.8584	.6748	.0692
166.2472	.3851	.1409	.6540	1.8602	.6745	.0647
178.4681	.3854	.1405	.6551	1.8620	.6742	.0605
191.5287	.3855	.1402	.6561	1.8637	.6739	.0566
200.7317	.3856	.1400	.6567	1.8649	.6737	.0541

MACH NO = 5.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0267	1.0457
.9152	.1409	.9225	1.0927	-.0426	1.0075	1.0036
1.2231	.1576	.8816	.8371	.0974	.9830	.9772
1.7377	.1799	.8190	.6507	.2841	.9503	.9360
2.4345	.1999	.7449	.5479	.4873	.9147	.8855
3.3047	.2143	.6663	.4934	.6944	.8785	.8296
4.4947	.2258	.5788	.4664	.9157	.8398	.7636
5.6902	.2337	.5088	.4624	1.0815	.8108	.7071
7.0188	.2412	.4467	.4701	1.2152	.7874	.6534
8.4816	.2493	.3927	.4846	1.3181	.7694	.6030
10.0922	.2583	.3459	.5027	1.3946	.7560	.5558
11.8701	.2685	.3056	.5223	1.4503	.7462	.5116
14.0351	.2808	.2680	.5435	1.4940	.7386	.4664
16.0152	.2916	.2414	.5597	1.5212	.7338	.4315
18.0708	.3020	.2196	.5736	1.5428	.7300	.4004
20.2044	.3119	.2014	.5852	1.5614	.7268	.3726
22.4269	.3211	.1862	.5948	1.5785	.7238	.3474
24.7572	.3297	.1733	.6029	1.5949	.7209	.3244
27.2168	.3375	.1622	.6096	1.6112	.7181	.3033
30.2112	.3457	.1513	.6159	1.6299	.7148	.2809
33.0103	.3522	.1431	.6206	1.6462	.7120	.2629
35.9922	.3580	.1360	.6245	1.6623	.7091	.2460
39.1713	.3633	.1298	.6279	1.6781	.7064	.2302
42.5618	.3680	.1244	.6308	1.6935	.7037	.2155
46.1770	.3722	.1197	.6333	1.7082	.7011	.2018
50.0301	.3760	.1156	.6356	1.7221	.6987	.1889
54.1349	.3794	.1121	.6376	1.7353	.6964	.1769
58.5065	.3824	.1090	.6395	1.7475	.6942	.1657
63.8508	.3854	.1059	.6414	1.7606	.6919	.1538
68.8528	.3878	.1036	.6429	1.7711	.6901	.1441
74.1796	.3898	.1016	.6444	1.7811	.6884	.1350
79.8529	.3915	.0998	.6456	1.7904	.6867	.1265
85.8955	.3930	.0982	.6468	1.7991	.6852	.1186
92.3327	.3943	.0969	.6479	1.8072	.6838	.1112
99.1917	.3953	.0957	.6489	1.8149	.6824	.1042
106.5022	.3962	.0946	.6498	1.8220	.6812	.0977
115.4505	.3969	.0936	.6508	1.8295	.6799	.0908
123.8379	.3974	.0928	.6517	1.8355	.6788	.0851
132.7840	.3978	.0920	.6524	1.8411	.6779	.0798
142.3279	.3981	.0914	.6532	1.8461	.6770	.0748
152.5116	.3982	.0908	.6539	1.8506	.6762	.0701
163.3801	.3983	.0903	.6546	1.8548	.6755	.0657
174.9818	.3983	.0899	.6553	1.8585	.6748	.0616
187.3688	.3982	.0894	.6560	1.8620	.6742	.0578
200.5972	.3980	.0891	.6566	1.8651	.6737	.0542

MACH NO = 10.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0250	1.0401
.9642	.1408	.8799	1.0379	-.0183	1.0032	.9993
1.4237	.1595	.8211	.7390	.1785	.9688	.9607
2.1606	.1777	.7383	.5646	.4255	.9255	.9047
3.1257	.1875	.6473	.4783	.6852	.8801	.8405
4.2759	.1906	.5596	.4380	.9311	.8371	.7749
5.5447	.1909	.4826	.4253	1.1370	.8011	.7136
6.8732	.1911	.4188	.4291	1.2927	.7738	.6589
8.2255	.1924	.3669	.4426	1.4014	.7548	.6113
9.5838	.1953	.3248	.4612	1.4712	.7426	.5699
10.9387	.1998	.2906	.4823	1.5115	.7355	.5338
12.1367	.2049	.2654	.5015	1.5294	.7314	.5055
13.4785	.2119	.2416	.5225	1.5356	.7313	.4772
14.9734	.2209	.2195	.5444	1.5320	.7319	.4492
16.4979	.2310	.2009	.5643	1.5229	.7335	.4238
18.0952	.2418	.1846	.5820	1.5130	.7353	.4001
19.8248	.2533	.1699	.5974	1.5056	.7365	.3772
21.7589	.2652	.1563	.6102	1.5038	.7369	.3546
23.9790	.2771	.1435	.6206	1.5093	.7359	.3318
26.5126	.2887	.1316	.6284	1.5222	.7336	.3090
29.2424	.2991	.1213	.6340	1.5400	.7305	.2878
32.1656	.3085	.1124	.6381	1.5602	.7270	.2681
35.2797	.3169	.1048	.6411	1.5815	.7233	.2498
38.5935	.3243	.0982	.6433	1.6033	.7195	.2329
42.1293	.3307	.0926	.6449	1.6255	.7156	.2173
45.9121	.3361	.0876	.6459	1.6480	.7116	.2027
49.9655	.3406	.0832	.6465	1.6703	.7077	.1891
54.3111	.3444	.0794	.6469	1.6918	.7040	.1764
58.9673	.3476	.0760	.6472	1.7121	.7004	.1646
63.9497	.3502	.0731	.6475	1.7310	.6971	.1536
69.2750	.3524	.0706	.6479	1.7480	.6941	.1433
74.9639	.3543	.0683	.6484	1.7632	.6915	.1338
81.0421	.3559	.0664	.6489	1.7768	.6891	.1249
87.5419	.3573	.0646	.6496	1.7887	.6870	.1166
94.5021	.3585	.0631	.6503	1.7993	.6852	.1089
101.9674	.3595	.0618	.6510	1.8088	.6835	.1017
109.9869	.3604	.0606	.6517	1.8173	.6820	.0949
118.6135	.3611	.0595	.6525	1.8250	.6807	.0885
127.9036	.3616	.0585	.6532	1.8321	.6794	.0826
137.9165	.3620	.0576	.6539	1.8385	.6783	.0770
148.7147	.3623	.0569	.6545	1.8444	.6773	.0718
160.3640	.3624	.0561	.6552	1.8499	.6763	.0669
171.6333	.3624	.0556	.6557	1.8543	.6755	.0628
185.0938	.3623	.0550	.6564	1.8587	.6748	.0584
201.1301	.3622	.0544	.6570	1.8630	.6740	.0540

MACH NO = 15.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0247	1.0391
1.0095	.1422	.8670	.9929	.0035	.9994	.9954
1.5546	.1619	.7988	.6914	.2280	.9601	.9503
2.3227	.1768	.7152	.5405	.4766	.9166	.8932
3.4199	.1835	.6166	.4580	.7624	.8666	.8227
4.6987	.1835	.5256	.4226	1.0219	.8212	.7533
5.9251	.1817	.4566	.4142	1.2097	.7883	.6970
7.3060	.1803	.3946	.4199	1.3624	.7616	.6429
8.6669	.1803	.3460	.4345	1.4635	.7439	.5972
9.8543	.1818	.3111	.4514	1.5200	.7340	.5623
11.1187	.1848	.2802	.4717	1.5546	.7280	.5294
12.2034	.1887	.2577	.4903	1.5675	.7257	.5040
13.3537	.1941	.2372	.5104	1.5680	.7256	.4797
14.6763	.2017	.2172	.5332	1.5567	.7276	.4545
15.8611	.2096	.2019	.5525	1.5404	.7305	.4340
17.1636	.2191	.1875	.5714	1.5210	.7339	.4136
18.5276	.2292	.1745	.5882	1.5034	.7369	.3941
19.8882	.2389	.1633	.6016	1.4915	.7390	.3765
21.5696	.2499	.1515	.6139	1.4854	.7401	.3567
23.5150	.2609	.1400	.6236	1.4884	.7396	.3363
25.5779	.2708	.1298	.6304	1.4987	.7378	.3170
28.2819	.2818	.1189	.6363	1.5170	.7346	.2949
31.1030	.2914	.1097	.6404	1.5375	.7310	.2749
34.3564	.3008	.1012	.6436	1.5612	.7268	.2550
37.8223	.3087	.0940	.6455	1.5864	.7224	.2367
41.2326	.3148	.0882	.6465	1.6112	.7181	.2211
45.2474	.3202	.0826	.6468	1.6390	.7132	.2051
49.5986	.3246	.0777	.6469	1.6662	.7085	.1903
53.8930	.3279	.0738	.6469	1.6896	.7044	.1776
58.9144	.3310	.0700	.6470	1.7128	.7003	.1647
64.2956	.3336	.0668	.6473	1.7332	.6967	.1529
69.5631	.3356	.0643	.6478	1.7497	.6938	.1428
75.7114	.3376	.0618	.6483	1.7655	.6911	.1326
81.7594	.3391	.0599	.6490	1.7783	.6888	.1239
88.8620	.3406	.0580	.6497	1.7907	.6867	.1151
96.5530	.3418	.0563	.6505	1.8017	.6847	.1068
104.1745	.3428	.0550	.6513	1.8108	.6832	.0997
113.1730	.3437	.0536	.6521	1.8197	.6816	.0924
122.9526	.3444	.0525	.6530	1.8277	.6802	.0857
132.6666	.3449	.0515	.6537	1.8343	.6790	.0799
144.1556	.3454	.0505	.6545	1.8409	.6779	.0739
156.6577	.3457	.0497	.6553	1.8466	.6769	.0684
169.0840	.3459	.0490	.6560	1.8513	.6761	.0637
183.7789	.3460	.0483	.6568	1.8556	.6753	.0588
201.1450	.3462	.0477	.6577	1.8593	.6747	.0540

MACH NO = 20.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1285	.9279	1.3710	-.1405	1.0246	1.0388
1.0091	.1419	.8644	.9933	.0034	.9994	.9954
1.5495	.1611	.7969	.6926	.2264	.9604	.9507
2.4075	.1765	.7044	.5295	.5025	.9121	.8873
3.5124	.1817	.6069	.4518	.7869	.8623	.8172
4.7818	.1807	.5182	.4181	1.0423	.8176	.7492
6.1194	.1780	.4447	.4099	1.2437	.7824	.6888
7.4585	.1761	.3864	.4160	1.3881	.7571	.6374
8.7607	.1757	.3407	.4301	1.4836	.7404	.5943
10.0029	.1768	.3049	.4478	1.5417	.7302	.5582
11.1725	.1792	.2767	.4668	1.5728	.7248	.5280
12.2667	.1827	.2542	.4860	1.5847	.7227	.5026
13.2925	.1873	.2359	.5046	1.5834	.7229	.4809
14.5461	.1943	.2167	.5276	1.5696	.7254	.4568
15.7443	.2023	.2011	.5487	1.5489	.7290	.4360
16.8378	.2105	.1886	.5663	1.5280	.7326	.4185
18.0542	.2198	.1766	.5834	1.5069	.7363	.4007
19.2568	.2289	.1662	.5972	1.4911	.7391	.3845
20.7170	.2392	.1552	.6100	1.4803	.7410	.3665
22.2688	.2488	.1451	.6197	1.4782	.7413	.3491
24.1885	.2589	.1344	.6277	1.4849	.7402	.3298
26.3033	.2684	.1246	.6335	1.4980	.7379	.3108
29.0907	.2791	.1140	.6388	1.5179	.7344	.2889
32.0223	.2887	.1051	.6426	1.5393	.7306	.2690
35.4035	.2978	.0969	.6453	1.5645	.7263	.2491
38.7545	.3047	.0902	.6465	1.5905	.7217	.2322
42.6932	.3107	.0839	.6468	1.6207	.7164	.2150
46.6493	.3152	.0787	.6468	1.6486	.7115	.2001
51.2897	.3191	.0739	.6466	1.6771	.7066	.1851
55.8921	.3222	.0700	.6467	1.7007	.7024	.1722
61.2234	.3251	.0664	.6469	1.7231	.6985	.1594
66.4820	.3273	.0635	.6473	1.7412	.6953	.1485
72.5918	.3295	.0607	.6479	1.7585	.6923	.1376
78.6677	.3311	.0585	.6485	1.7726	.6898	.1282
85.7883	.3327	.0564	.6493	1.7861	.6875	.1187
92.9166	.3340	.0547	.6501	1.7971	.6855	.1106
101.3026	.3352	.0531	.6510	1.8077	.6837	.1023
109.7140	.3361	.0517	.6518	1.8163	.6822	.0951
119.6229	.3370	.0504	.6528	1.8247	.6807	.0879
129.5751	.3376	.0493	.6536	1.8317	.6795	.0816
141.3146	.3382	.0483	.6545	1.8384	.6783	.0753
153.1154	.3386	.0474	.6553	1.8438	.6774	.0699
167.0356	.3390	.0466	.6562	1.8488	.6765	.0644
181.0158	.3394	.0460	.6571	1.8526	.6758	.0597
200.1377	.3398	.0453	.6582	1.8562	.6752	.0543

MACH NO = 25.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0245	1.0386
1.0582	.1439	.8570	.9498	.1263	.9954	.9912
1.6202	.1627	.7876	.6707	.2521	.9559	.9451
2.5007	.1768	.6943	.5190	.5299	.9073	.8810
3.6194	.1809	.5976	.4463	.8128	.8578	.8110
4.8890	.1792	.5107	.4151	1.0638	.8139	.7440
6.2146	.1763	.4391	.4078	1.2603	.7795	.6849
7.5325	.1741	.3826	.4141	1.4007	.7549	.6348
8.8058	.1735	.3383	.4278	1.4935	.7387	.5929
10.0125	.1743	.3037	.4450	1.5503	.7287	.5579
11.1406	.1764	.2764	.4633	1.5810	.7234	.5288
12.1885	.1795	.2547	.4817	1.5930	.7213	.5044
13.2574	.1840	.2355	.5015	1.5915	.7215	.4816
14.4310	.1904	.2173	.5236	1.5775	.7240	.4590
15.5405	.1978	.2026	.5440	1.5567	.7276	.4394
16.6230	.2059	.1900	.5625	1.5337	.7316	.4218
17.7187	.2144	.1788	.5791	1.5118	.7355	.4054
18.8767	.2235	.1684	.5937	1.4932	.7387	.3894
20.1586	.2329	.1583	.6065	1.4800	.7410	.3731
21.6328	.2425	.1481	.6171	1.4746	.7420	.3560
23.3048	.2519	.1382	.6252	1.4779	.7414	.3384
25.2736	.2613	.1282	.6315	1.4889	.7395	.3197
27.6961	.2712	.1181	.6369	1.5057	.7365	.2995
30.5454	.2814	.1084	.6413	1.5264	.7329	.2787
33.7222	.2908	.0998	.6446	1.5497	.7288	.2586
37.1220	.2987	.0924	.6463	1.5764	.7242	.2401
40.8152	.3050	.0858	.6468	1.6063	.7189	.2229
44.8512	.3100	.0800	.6467	1.6368	.7136	.2066
49.5929	.3144	.0745	.6465	1.6681	.7081	.1903
54.3448	.3178	.0702	.6465	1.6940	.7036	.1763
59.4252	.3207	.0664	.6467	1.7167	.6996	.1635
64.8532	.3231	.0632	.6471	1.7365	.6962	.1517
70.6803	.3252	.0604	.6476	1.7540	.6931	.1408
76.9705	.3271	.0580	.6483	1.7694	.6904	.1307
83.7916	.3287	.0559	.6490	1.7830	.6880	.1213
91.2063	.3301	.0540	.6499	1.7949	.6859	.1124
99.2732	.3313	.0523	.6508	1.8053	.6841	.1041
108.0542	.3324	.0508	.6517	1.8145	.6825	.0964
117.6190	.3333	.0495	.6527	1.8227	.6811	.0892
128.0471	.3340	.0483	.6536	1.8301	.6798	.0825
139.4246	.3347	.0472	.6545	1.8366	.6786	.0763
151.8403	.3352	.0463	.6554	1.8422	.6777	.0704
165.3829	.3357	.0455	.6564	1.8469	.6768	.0650
180.1405	.3362	.0449	.6573	1.8508	.6762	.0600
200.1008	.3369	.0442	.6586	1.8542	.6756	.0543

MACH NO = 30.00 CONE ANGLE = 5.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0245	1.0385
1.3570	.1439	.8564	.9500	.0262	.9954	.9912
1.6186	.1625	.7872	.6710	.2517	.9560	.9452
2.4967	.1765	.6941	.5190	.5291	.9074	.8912
3.7328	.1804	.5884	.4413	.8389	.8572	.8046
5.0058	.1783	.5031	.4128	1.0853	.8101	.7384
6.3246	.1752	.4374	.4067	1.2765	.7766	.6804
7.6288	.1730	.3794	.4135	1.4124	.7529	.6314
9.0056	.1723	.3316	.4289	1.5089	.7360	.5868
10.1807	.1731	.2988	.4459	1.5608	.7269	.5534
11.2741	.1752	.2729	.4639	1.5882	.7221	.5256
12.2856	.1782	.2522	.4819	1.5983	.7203	.5022
13.3139	.1826	.2341	.5012	1.5955	.7218	.4805
14.4390	.1889	.2167	.5227	1.5808	.7234	.4588
15.4983	.1958	.2025	.5426	1.5598	.7271	.4401
16.6051	.2041	.1896	.5620	1.5350	.7314	.4221
17.6427	.2123	.1790	.5781	1.5130	.7353	.4066
18.8156	.2216	.1694	.5934	1.4927	.7388	.3902
20.0222	.2305	.1588	.6058	1.4792	.7412	.3748
21.4053	.2393	.1491	.6162	1.4728	.7423	.3585
23.0955	.2494	.1388	.6247	1.4755	.7418	.3405
24.9423	.2583	.1292	.6319	1.4856	.7400	.3227
27.2076	.2678	.1194	.6362	1.5012	.7373	.3033
30.1131	.2784	.1093	.6410	1.5222	.7337	.2816
33.1651	.2878	.1007	.6444	1.5444	.7298	.2619
36.4616	.2957	.0932	.6463	1.5704	.7252	.2435
40.3292	.3025	.0860	.6468	1.6024	.7196	.2250
44.2690	.3075	.0802	.6466	1.6331	.7143	.2088
48.8993	.3119	.0747	.6464	1.6644	.7088	.1925
53.5331	.3153	.0703	.6463	1.6904	.7042	.1786
58.4784	.3182	.0665	.6465	1.7131	.7002	.1658
64.1740	.3208	.0630	.6470	1.7345	.6965	.1531
69.8624	.3229	.0602	.6475	1.7520	.6934	.1423
76.0010	.3248	.0577	.6481	1.7675	.6907	.1322
83.1910	.3265	.0554	.6489	1.7822	.6882	.1220
90.4673	.3279	.0535	.6498	1.7939	.6861	.1132
98.3772	.3291	.0518	.6507	1.8042	.6843	.1050
107.6714	.3303	.0502	.6517	1.8139	.6826	.0967
117.0943	.3312	.0489	.6527	1.8220	.6812	.0896
128.1874	.3321	.0476	.6537	1.8297	.6798	.0824
139.4547	.3328	.0466	.6546	1.8360	.6787	.0762
151.7401	.3334	.0457	.6556	1.8414	.6778	.0705
166.2048	.3339	.0449	.6566	1.8462	.6770	.0647
180.8702	.3345	.0442	.6576	1.8497	.6763	.0597
200.6746	.3352	.0435	.6589	1.8528	.6758	.0541

MACH NO = 3.50 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0351	1.0542
.8518	.1395	.9807	1.1740	-.0749	1.0158	1.0112
1.1103	.1567	.9369	.9136	.0472	.9901	.9832
1.5278	.1815	.8750	.7199	.2017	.9576	.9425
2.0938	.2074	.8019	.6100	.3644	.9234	.8925
2.8222	.2295	.7226	.5485	.5322	.8881	.8354
3.7155	.2472	.6431	.5172	.6948	.8539	.7746
4.7693	.2612	.5683	.5050	.8421	.8230	.7134
5.9759	.2730	.5013	.5052	.9672	.7967	.6542
7.3277	.2834	.4435	.5128	1.0684	.7754	.5986
8.8183	.2930	.3946	.5240	1.1485	.7586	.5473
10.2037	.3005	.3592	.5346	1.2034	.7470	.5069
11.9462	.3085	.3246	.5471	1.2547	.7363	.4638
13.8239	.3160	.2950	.5591	1.2951	.7278	.4249
15.5470	.3220	.2754	.5686	1.3232	.7218	.3946
17.3785	.3275	.2580	.5773	1.3470	.7168	.3667
19.6590	.3333	.2409	.5863	1.3707	.7119	.3371
21.7444	.3378	.2285	.5932	1.3882	.7082	.3139
23.9596	.3418	.2179	.5994	1.4037	.7049	.2925
26.7226	.3460	.2074	.6059	1.4197	.7016	.2696
29.2586	.3492	.1998	.6108	1.4321	.6990	.2515
31.9641	.3520	.1932	.6153	1.4433	.6966	.2347
35.3542	.3550	.1866	.6200	1.4552	.6941	.2166
38.4769	.3572	.1817	.6236	1.4644	.6922	.2022
41.8158	.3592	.1775	.6270	1.4728	.6904	.1888
46.0049	.3613	.1733	.6305	1.4817	.6885	.1743
49.8661	.3628	.1702	.6333	1.4885	.6871	.1628
53.9950	.3642	.1676	.6360	1.4947	.6858	.1521
59.1747	.3656	.1649	.6388	1.5012	.6844	.1405
63.9473	.3667	.1630	.6410	1.5062	.6834	.1312
69.0488	.3677	.1613	.6430	1.5107	.6824	.1226
75.4456	.3687	.1596	.6453	1.5153	.6815	.1132
81.3371	.3694	.1584	.6471	1.5189	.6807	.1058
87.6329	.3701	.1573	.6487	1.5222	.6800	.0989
95.5261	.3708	.1563	.6505	1.5257	.6793	.0914
102.7959	.3712	.1555	.6519	1.5284	.6787	.0854
110.5655	.3716	.1548	.6532	1.5310	.6782	.0799
120.3089	.3720	.1541	.6546	1.5337	.6776	.0738
129.2856	.3722	.1536	.6557	1.5359	.6771	.0690
138.8828	.3724	.1532	.6568	1.5380	.6767	.0645
150.9232	.3725	.1528	.6579	1.5402	.6762	.0597
162.0213	.3726	.1524	.6587	1.5419	.6759	.0558
173.8917	.3726	.1521	.6596	1.5435	.6755	.0521
188.7906	.3725	.1518	.6604	1.5452	.6752	.0482
200.1744	.3724	.1516	.6610	1.5463	.6749	.0456

MACH NO = 5.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0321	1.0457
.8763	.1386	.9289	1.1412	-.0624	1.0131	1.0077
1.1632	.1542	.8830	.8764	.0703	.9852	.9778
1.6439	.1750	.8140	.6800	.2451	.9485	.9318
2.2719	.1932	.7362	.5742	.4246	.9107	.8778
3.0446	.2063	.6558	.5174	.6020	.8735	.8194
3.9509	.2153	.5786	.4903	.7653	.8391	.7601
4.9740	.2225	.5086	.4825	.9043	.8099	.7026
6.0993	.2293	.4473	.4871	1.0147	.7867	.6487
7.3192	.2365	.3948	.4989	1.0984	.7691	.5989
8.6364	.2444	.3501	.5144	1.1597	.7562	.5531
10.0626	.2532	.3120	.5315	1.2039	.7469	.5107
11.6141	.2628	.2795	.5487	1.2357	.7402	.4715
13.5494	.2744	.2482	.5669	1.2622	.7347	.4302
15.5816	.2857	.2233	.5824	1.2820	.7305	.3940
17.4297	.2951	.2056	.5935	1.2964	.7275	.3660
19.6377	.3051	.1890	.6041	1.3114	.7243	.3373
21.6728	.3132	.1769	.6116	1.3242	.7216	.3146
24.1404	.3217	.1651	.6187	1.3389	.7186	.2909
26.4412	.3284	.1564	.6237	1.3519	.7158	.2718
29.2517	.3353	.1478	.6285	1.3669	.7127	.2516
31.8848	.3407	.1414	.6320	1.3800	.7099	.2352
35.1104	.3462	.1351	.6353	1.3946	.7068	.2178
38.1362	.3504	.1303	.6378	1.4069	.7043	.2037
41.8436	.3547	.1256	.6403	1.4204	.7014	.1887
45.8355	.3585	.1217	.6424	1.4331	.6988	.1749
49.5781	.3614	.1187	.6441	1.4436	.6966	.1636
54.1617	.3642	.1157	.6458	1.4548	.6942	.1517
58.4593	.3664	.1135	.6472	1.4640	.6923	.1420
63.7227	.3685	.1113	.6485	1.4739	.6902	.1316
68.6576	.3700	.1096	.6496	1.4819	.6885	.1232
74.7021	.3714	.1079	.6508	1.4905	.6867	.1143
80.3706	.3725	.1067	.6517	1.4974	.6852	.1070
87.3161	.3734	.1054	.6527	1.5047	.6837	.0992
93.8326	.3741	.1044	.6535	1.5104	.6825	.0929
101.8211	.3746	.1035	.6544	1.5164	.6812	.0862
110.4319	.3751	.1026	.6553	1.5217	.6801	.0800
118.5170	.3753	.1020	.6561	1.5259	.6792	.0749
128.4363	.3755	.1013	.6569	1.5301	.6784	.0694
137.7546	.3756	.1008	.6576	1.5334	.6777	.0650
149.1925	.3757	.1003	.6584	1.5368	.6770	.0603
159.9424	.3756	.0999	.6591	1.5394	.6764	.0565
173.1431	.3756	.0995	.6598	1.5420	.6759	.0524
185.5549	.3754	.0992	.6604	1.5440	.6754	.0490
200.8017	.3753	.0988	.6611	1.5461	.6750	.0454

MACH NO = 10.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0301	1.0401
.9650	.1406	.8782	1.0377	-.0182	1.0038	.9982
1.4024	.1578	.8123	.7520	.1659	.9651	.9544
1.9953	.1723	.7343	.5977	.3615	.9240	.9008
2.8453	.1815	.6408	.5071	.5846	.8771	.8337
3.8516	.1844	.5521	.4638	.7912	.8337	.7662
4.8293	.1848	.4831	.4496	.9439	.8016	.7103
5.9734	.1852	.4187	.4503	1.0743	.7742	.6544
7.1317	.1865	.3669	.4612	1.1644	.7552	.6061
8.1587	.1888	.3295	.4756	1.2169	.7442	.5689
9.3025	.1928	.2953	.4942	1.2527	.7367	.5324
10.3050	.1973	.2702	.5112	1.2696	.7331	.5042
11.4172	.2033	.2469	.5300	1.2773	.7315	.4761
12.7652	.2118	.2235	.5516	1.2764	.7317	.4460
14.0014	.2203	.2057	.5695	1.2706	.7329	.4216
15.4107	.2305	.1890	.5871	1.2625	.7346	.3968
16.7797	.2404	.1754	.6011	1.2563	.7359	.3754
18.4412	.2516	.1618	.6142	1.2531	.7366	.3523
20.1558	.2620	.1502	.6240	1.2550	.7362	.3313
22.3421	.2734	.1382	.6326	1.2633	.7344	.3078
24.6573	.2835	.1280	.6385	1.2752	.7317	.2864
27.4609	.2936	.1182	.6432	1.2939	.7290	.2641
30.1986	.3018	.1107	.6461	1.3114	.7248	.2454
33.4284	.3096	.1036	.6484	1.3315	.7201	.2266
36.5817	.3156	.0982	.6498	1.3500	.7152	.2107
40.3161	.3213	.0929	.6507	1.3705	.7119	.1946
43.9763	.3255	.0888	.6512	1.3887	.7081	.1811
48.3191	.3293	.0849	.6516	1.4079	.7040	.1673
52.5745	.3322	.0818	.6518	1.4242	.7006	.1556
57.6152	.3349	.0789	.6521	1.4406	.6972	.1438
62.5455	.3368	.0766	.6525	1.4540	.6944	.1338
68.3801	.3387	.0744	.6530	1.4670	.6916	.1236
74.0889	.3401	.0727	.6535	1.4775	.6894	.1151
80.8556	.3414	.0710	.6542	1.4876	.6873	.1064
87.4925	.3425	.0697	.6548	1.4956	.6856	.0990
95.3806	.3434	.0684	.6556	1.5034	.6840	.0915
103.1366	.3441	.0674	.6563	1.5097	.6826	.0852
112.3737	.3447	.0664	.6571	1.5159	.6813	.0787
122.4194	.3452	.0655	.6578	1.5213	.6802	.0726
132.3138	.3455	.0648	.6585	1.5257	.6793	.0675
144.1122	.3458	.0641	.6593	1.5300	.6784	.0623
155.7371	.3460	.0635	.6599	1.5334	.6777	.0579
169.5959	.3461	.0630	.6607	1.5365	.6770	.0534
183.2456	.3462	.0626	.6614	1.5390	.6765	.0496
201.0552	.3463	.0622	.6622	1.5413	.6760	.0454

MACH NO = 15.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0297	1.0391
1.0105	.1420	.8641	.9931	.0035	.9993	.9934
1.4566	.1579	.7980	.7301	.1866	.9608	.9492
2.1383	.1716	.7106	.5731	.4057	.9147	.8888
2.9987	.1777	.6195	.4924	.6261	.8684	.8227
3.9901	.1783	.5350	.4533	.8265	.8263	.7577
5.1688	.1768	.4560	.4389	1.0045	.7888	.6927
6.2455	.1758	.3991	.4412	1.1208	.7644	.6423
7.3078	.1759	.3535	.4517	1.2007	.7476	.5993
8.3394	.1772	.3172	.4665	1.2519	.7368	.5628
9.3309	.1797	.2879	.4831	1.2819	.7305	.5316
10.2778	.1831	.2642	.5002	1.2969	.7274	.5049
11.2796	.1878	.2429	.5187	1.3014	.7264	.4794
12.4252	.1946	.2222	.5397	1.2961	.7276	.4532
13.5746	.2022	.2054	.5591	1.2845	.7300	.4305
14.6401	.2106	.1911	.5766	1.2707	.7329	.4100
15.7789	.2195	.1786	.5922	1.2574	.7357	.3908
16.9954	.2288	.1671	.6058	1.2469	.7379	.3722
18.4665	.2393	.1552	.6183	1.2404	.7392	.3520
20.0266	.2491	.1447	.6277	1.2406	.7392	.3327
21.8058	.2588	.1346	.6349	1.2468	.7379	.3133
23.9067	.2685	.1248	.6406	1.2596	.7354	.2930
26.4372	.2783	.1154	.6451	1.2751	.7320	.2718
29.3296	.2876	.1068	.6483	1.2947	.7278	.2511
32.4342	.2956	.0995	.6504	1.3156	.7234	.2321
36.0707	.3027	.0928	.6515	1.3396	.7184	.2131
39.7220	.3080	.0875	.6519	1.3623	.7136	.1970
43.6870	.3123	.0828	.6519	1.3844	.7090	.1821
47.9841	.3159	.0788	.6519	1.4050	.7047	.1682
52.6249	.3188	.0753	.6520	1.4236	.7008	.1555
57.6248	.3213	.0724	.6523	1.4400	.6973	.1437
63.4775	.3236	.0696	.6527	1.4556	.6940	.1321
69.3253	.3253	.0674	.6532	1.4682	.6914	.1221
75.6522	.3268	.0655	.6538	1.4793	.6890	.1130
82.5141	.3281	.0639	.6545	1.4889	.6870	.1045
89.9693	.3292	.0624	.6553	1.4974	.6852	.0966
98.0777	.3301	.0611	.6560	1.5049	.6837	.0892
106.9016	.3308	.0600	.6568	1.5114	.6823	.0824
117.3450	.3315	.0589	.6577	1.5175	.6810	.0756
127.8775	.3321	.0581	.6586	1.5223	.6800	.0697
139.7453	.3325	.0573	.6594	1.5265	.6791	.0643
151.8298	.3329	.0567	.6602	1.5299	.6784	.0593
165.4163	.3333	.0561	.6611	1.5328	.6778	.0547
180.1943	.3337	.0557	.6620	1.5351	.6773	.0504
200.4852	.3342	.0552	.6631	1.5371	.6769	.0455

MACH NO = 20.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1285	.9279	1.3710	-.1405	1.0295	1.0388
1.0100	.1417	.8616	.9935	.0033	.9993	.9935
1.4523	.1572	.7961	.7313	.1853	.9610	.9496
2.2137	.1714	.6994	.5617	.4282	.9100	.8826
3.0806	.1762	.6093	.4858	.6470	.8640	.8169
4.1810	.1758	.5184	.4463	.8639	.8184	.7464
5.2261	.1738	.4503	.4351	1.0183	.7859	.6898
6.3904	.1722	.3900	.4379	1.1424	.7599	.6361
7.4129	.1720	.3472	.4483	1.2174	.7441	.5954
8.5020	.1731	.3097	.4643	1.2695	.7331	.5574
9.4307	.1751	.2829	.4801	1.2961	.7276	.5287
10.3076	.1780	.2612	.4961	1.3091	.7248	.5041
11.2245	.1821	.2416	.5136	1.3124	.7241	.4807
12.3424	.1884	.2212	.5351	1.3054	.7256	.4550
13.4049	.1957	.2049	.5550	1.2917	.7285	.4330
14.4457	.2037	.1912	.5729	1.2755	.7319	.4134
15.5010	.2121	.1792	.5888	1.2597	.7352	.3953
16.6139	.2210	.1682	.6028	1.2466	.7380	.3778
17.8385	.2302	.1578	.6149	1.2377	.7398	.3603
19.2330	.2397	.1476	.6249	1.2345	.7405	.3422
20.8074	.2489	.1379	.6327	1.2378	.7398	.3239
22.6466	.2582	.1283	.6387	1.2469	.7379	.3048
24.8833	.2678	.1188	.6437	1.2609	.7349	.2844
27.5644	.2774	.1097	.6476	1.2789	.7312	.2633
30.6266	.2864	.1014	.6503	1.2999	.7267	.2428
33.9394	.2939	.0944	.6517	1.3230	.7219	.2238
37.5461	.2999	.0883	.6521	1.3477	.7167	.2063
41.4870	.3047	.0830	.6520	1.3721	.7116	.1901
45.7822	.3086	.0784	.6519	1.3950	.7068	.1751
50.4355	.3119	.0745	.6519	1.4156	.7024	.1613
55.4578	.3146	.0712	.6521	1.4336	.6986	.1486
60.8774	.3169	.0684	.6525	1.4494	.6953	.1370
66.7431	.3188	.0659	.6530	1.4632	.6924	.1263
73.1153	.3204	.0638	.6536	1.4753	.6899	.1165
80.0586	.3219	.0619	.6543	1.4858	.6877	.1074
87.6370	.3230	.0603	.6550	1.4949	.6858	.0989
95.9147	.3240	.0589	.6559	1.5028	.6841	.0911
104.9591	.3249	.0577	.6568	1.5097	.6827	.0838
114.8438	.3256	.0566	.6577	1.5155	.6814	.0771
125.6495	.3263	.0557	.6586	1.5205	.6804	.0709
137.4628	.3269	.0549	.6595	1.5247	.6795	.0652
150.3744	.3275	.0542	.6605	1.5281	.6788	.0599
164.4788	.3280	.0536	.6614	1.5308	.6782	.0550
179.8739	.3286	.0531	.6624	1.5329	.6778	.0505
200.7452	.3293	.0527	.6636	1.5347	.6774	.0455

MACH NO = 25.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0295	1.0386
1.0098	.1416	.8606	.9937	.0032	.9993	.9935
1.5157	.1587	.7863	.7086	.2084	.9562	.9437
2.2081	.1708	.6991	.5619	.4271	.9102	.8830
3.1742	.1755	.5997	.4797	.6692	.8593	.8104
4.2759	.1745	.5106	.4431	.8820	.8146	.7408
5.3129	.1723	.4444	.4330	1.0325	.7830	.6855
6.4610	.1705	.3858	.4363	1.1529	.7576	.6331
7.5728	.1701	.3402	.4481	1.2319	.7410	.5895
8.5265	.1710	.3079	.4622	1.2762	.7317	.5566
9.5245	.1731	.2794	.4793	1.3041	.7259	.5259
10.3653	.1758	.2588	.4949	1.3156	.7234	.5026
11.2399	.1796	.2402	.5118	1.3180	.7229	.4804
12.3796	.1860	.2196	.5342	1.3095	.7247	.4542
13.3750	.1928	.2043	.5534	1.2951	.7278	.4336
14.3443	.2003	.1914	.5708	1.2785	.7313	.4153
15.3941	.2088	.1793	.5874	1.2611	.7349	.3971
16.4143	.2171	.1691	.6010	1.2473	.7378	.3809
17.6166	.2264	.1586	.6136	1.2366	.7401	.3634
18.8848	.2353	.1490	.6235	1.2322	.7410	.3466
20.4193	.2446	.1391	.6317	1.2343	.7405	.3282
22.0720	.2531	.1301	.6376	1.2419	.7389	.3105
24.2296	.2628	.1204	.6429	1.2550	.7362	.2901
26.6473	.2720	.1116	.6469	1.2710	.7328	.2702
29.6124	.2814	.1030	.6500	1.2912	.7286	.2492
32.7570	.2891	.0957	.6517	1.3135	.7239	.2302
36.1755	.2953	.0894	.6522	1.3379	.7188	.2126
40.2116	.3007	.0835	.6521	1.3645	.7132	.1951
44.3111	.3047	.0788	.6519	1.3879	.7083	.1799
49.1034	.3082	.0745	.6518	1.4104	.7035	.1650
53.9091	.3110	.0711	.6520	1.4287	.6997	.1523
59.4971	.3135	.0679	.6523	1.4459	.6961	.1398
65.1196	.3154	.0654	.6528	1.4598	.6931	.1291
71.7107	.3172	.0631	.6534	1.4730	.6904	.1185
78.3941	.3186	.0612	.6541	1.4836	.6881	.1094
86.2670	.3199	.0594	.6549	1.4934	.6861	.1003
94.2667	.3209	.0580	.6557	1.5013	.6844	.0925
102.9935	.3218	.0567	.6566	1.5080	.6830	.0853
113.2837	.3227	.0555	.6576	1.5143	.6817	.0781
123.7485	.3234	.0546	.6586	1.5192	.6807	.0719
136.0932	.3241	.0537	.6596	1.5235	.6797	.0658
148.6444	.3248	.0530	.6606	1.5268	.6791	.0605
163.4354	.3255	.0524	.6616	1.5296	.6785	.0553
178.4493	.3261	.0520	.6626	1.5315	.6781	.0509
200.1025	.3270	.0515	.6639	1.5331	.6777	.0456

MACH NO = 30.00 CONE ANGLE = 6.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0294	1.0385
1.0097	.1415	.9600	.9938	.0031	.9993	.9935
1.5144	.1585	.7858	.7089	.2080	.9563	.9438
2.2050	.1705	.6988	.5620	.4265	.9103	.8833
3.1678	.1751	.5997	.4795	.6684	.8595	.8108
4.2648	.1739	.5109	.4426	.8814	.8147	.7415
5.2951	.1716	.4447	.4321	1.0321	.7830	.6863
6.4364	.1696	.3863	.4349	1.1532	.7576	.6342
7.5385	.1691	.3409	.4463	1.2329	.7408	.5908
8.4820	.1698	.3087	.4601	1.2777	.7314	.5581
9.4670	.1716	.2803	.4769	1.3063	.7254	.5276
10.2945	.1741	.2599	.4922	1.3194	.7229	.5044
11.1527	.1777	.2414	.5089	1.3213	.7222	.4825
12.2648	.1838	.2209	.5311	1.3134	.7239	.4567
13.3041	.1908	.2047	.5515	1.2980	.7272	.4350
14.2357	.1980	.1922	.5686	1.2812	.7307	.4173
15.2375	.2062	.1804	.5851	1.2635	.7344	.3997
16.2792	.2148	.1698	.5996	1.2481	.7376	.3829
17.4134	.2237	.1597	.6121	1.2367	.7400	.3662
18.7008	.2329	.1498	.6227	1.2311	.7412	.3489
20.0373	.2412	.1409	.6303	1.2320	.7410	.3326
21.6906	.2500	.1315	.6367	1.2389	.7396	.3144
23.6883	.2592	.1221	.6420	1.2508	.7371	.2950
26.1064	.2688	.1128	.6464	1.2665	.7338	.2744
28.8325	.2778	.1045	.6496	1.2849	.7299	.2544
31.7968	.2857	.0972	.6516	1.3058	.7255	.2357
35.3488	.2926	.0903	.6522	1.3318	.7201	.2167
39.2548	.2981	.0842	.6521	1.3586	.7144	.1990
43.5443	.3024	.0790	.6518	1.3840	.7091	.1826
48.2128	.3060	.0746	.6517	1.4067	.7043	.1676
52.8885	.3089	.0711	.6519	1.4252	.7004	.1548
58.3169	.3114	.0679	.6522	1.4427	.6967	.1423
64.2095	.3135	.0651	.6526	1.4579	.6935	.1307
70.6394	.3153	.0628	.6532	1.4712	.6907	.1201
77.1534	.3167	.0608	.6539	1.4819	.6885	.1110
84.8221	.3180	.0590	.6547	1.4918	.6864	.1019
93.2345	.3192	.0575	.6557	1.5003	.6846	.0935
102.4640	.3202	.0561	.6566	1.5075	.6831	.0857
112.5938	.3211	.0549	.6576	1.5136	.6818	.0785
122.8867	.3218	.0540	.6586	1.5184	.6808	.0724
135.0184	.3226	.0531	.6596	1.5227	.6799	.0663
148.3346	.3233	.0524	.6607	1.5261	.6792	.0606
162.9392	.3240	.0518	.6617	1.5288	.6786	.0555
178.9394	.3248	.0513	.6628	1.5307	.6782	.0507
200.4069	.3258	.0508	.6642	1.5322	.6779	.0455

MACH NO = 3.50 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0410	1.0542
.8518	.1395	.9807	1.1740	-.0749	1.0184	1.0112
1.1097	.1566	.9319	.9168	.0452	.9889	.9794
1.4553	.1770	.8733	.7490	.1718	.9578	.9404
1.9881	.2015	.7951	.6331	.3231	.9207	.8859
2.5545	.2198	.7238	.5760	.4532	.8887	.8336
3.3779	.2369	.6414	.5395	.5986	.8530	.7695
4.1915	.2487	.5753	.5256	.7105	.9255	.7146
5.2684	.2604	.5065	.5226	.8211	.7984	.6529
6.2938	.2694	.4555	.5272	.8977	.7795	.6033
7.6021	.2791	.4051	.5368	.9684	.7622	.5500
8.8149	.2868	.3689	.5465	1.0161	.7505	.5084
10.3340	.2952	.3339	.5582	1.0600	.7397	.4643
11.9695	.3029	.3053	.5695	1.0944	.7313	.4247
13.4648	.3090	.2849	.5784	1.1182	.7254	.3940
15.3244	.3156	.2650	.5879	1.1413	.7197	.3615
17.3150	.3215	.2486	.5964	1.1607	.7159	.3321
19.4468	.3268	.2351	.6037	1.1774	.7109	.3056
21.3969	.3309	.2253	.6093	1.1901	.7077	.2847
23.8330	.3351	.2156	.6151	1.2034	.7045	.2624
26.4509	.3388	.2075	.6203	1.2153	.7016	.2420
29.2787	.3421	.2007	.6248	1.2260	.6989	.2232
31.8820	.3446	.1957	.6283	1.2344	.6969	.2084
35.1464	.3472	.1907	.6321	1.2434	.6947	.1923
38.6755	.3495	.1865	.6354	1.2514	.6927	.1775
41.9278	.3512	.1834	.6381	1.2577	.6912	.1658
46.0081	.3530	.1803	.6409	1.2643	.6895	.1531
50.4200	.3545	.1777	.6435	1.2702	.6881	.1413
55.1901	.3558	.1755	.6459	1.2756	.6868	.1305
59.5854	.3568	.1739	.6477	1.2797	.6857	.1219
65.6982	.3579	.1723	.6497	1.2841	.6847	.1126
71.0575	.3587	.1709	.6516	1.2882	.6837	.1041
77.4996	.3594	.1698	.6533	1.2918	.6828	.0961
83.4356	.3599	.1689	.6546	1.2947	.6821	.0899
90.8819	.3603	.1681	.6560	1.2978	.6813	.0830
98.9336	.3607	.1674	.6574	1.3006	.6806	.0767
106.3549	.3609	.1668	.6584	1.3027	.6801	.0717
115.6679	.3610	.1663	.6596	1.3050	.6795	.0663
125.7424	.3611	.1658	.6606	1.3071	.6790	.0613
136.6424	.3612	.1654	.6616	1.3099	.6786	.0566
146.6949	.3612	.1651	.6624	1.3103	.6782	.0529
159.3166	.3611	.1648	.6633	1.3118	.6779	.0489
172.9777	.3611	.1645	.6641	1.3130	.6776	.0452
187.7651	.3610	.1643	.6649	1.3141	.6773	.0418
201.4075	.3608	.1641	.6656	1.3149	.6771	.0390

MACH NO = 5.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0375	1.0457
.8763	.1386	.9289	1.1412	-.0624	1.0153	1.0077
1.1621	.1538	.8765	.8803	.0677	.9834	.9733
1.6166	.1729	.8027	.6938	.2285	.9439	.9232
2.1925	.1890	.7228	.5927	.3870	.9050	.8666
2.8901	.2005	.6424	.5375	.5391	.8676	.8067
3.6975	.2086	.5667	.5107	.6758	.8341	.7470
4.5978	.2152	.4991	.5023	.7895	.8061	.6900
5.5755	.2215	.4408	.5057	.8781	.7844	.6372
6.6213	.2283	.3914	.5159	.9441	.7682	.5890
7.7340	.2356	.3497	.5295	.9919	.7564	.5452
8.7457	.2425	.3192	.5423	1.0220	.7490	.5106
10.0029	.2511	.2886	.5574	1.0477	.7427	.4733
11.5560	.2615	.2591	.5738	1.0690	.7375	.4341
13.2536	.2722	.2344	.5885	1.0855	.7334	.3981
15.0903	.2827	.2139	.6011	1.0994	.7300	.3653
17.0166	.2925	.1974	.6113	1.1120	.7269	.3362
19.0528	.3015	.1839	.6195	1.1243	.7239	.3102
21.2214	.3096	.1726	.6261	1.1357	.7209	.2865
23.5425	.3170	.1632	.6314	1.1494	.7177	.2649
26.0354	.3235	.1552	.6356	1.1622	.7146	.2450
28.7184	.3292	.1484	.6391	1.1750	.7115	.2267
31.6091	.3343	.1427	.6419	1.1874	.7084	.2098
34.7249	.3387	.1378	.6443	1.1995	.7055	.1942
38.0831	.3425	.1337	.6464	1.2109	.7026	.1798
41.7025	.3458	.1302	.6481	1.2217	.7000	.1665
45.6036	.3486	.1272	.6497	1.2319	.6975	.1542
49.8086	.3510	.1246	.6510	1.2415	.6951	.1428
54.3411	.3529	.1224	.6522	1.2503	.6930	.1323
59.2264	.3546	.1205	.6533	1.2585	.6909	.1226
64.4925	.3559	.1189	.6544	1.2660	.6891	.1136
70.1698	.3569	.1174	.6553	1.2729	.6874	.1053
76.2916	.3577	.1162	.6563	1.2790	.6859	.0975
82.8943	.3584	.1152	.6572	1.2844	.6846	.0904
90.0175	.3589	.1143	.6580	1.2893	.6834	.0838
97.7043	.3593	.1135	.6589	1.2935	.6823	.0776
106.0016	.3596	.1128	.6598	1.2973	.6814	.0719
114.9606	.3597	.1122	.6606	1.3006	.6806	.0667
124.6363	.3599	.1117	.6614	1.3034	.6799	.0618
135.0886	.3599	.1112	.6622	1.3059	.6793	.0572
146.3819	.3599	.1108	.6630	1.3081	.6788	.0530
158.5858	.3599	.1104	.6637	1.3100	.6783	.0491
170.0702	.3598	.1101	.6643	1.3114	.6780	.0459
184.1885	.3597	.1098	.6650	1.3128	.6776	.0426
201.4414	.3595	.1095	.6657	1.3141	.6773	.0390

MACH NO = 10.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0351	1.0401
.9661	.1407	.9761	1.0375	-.0180	1.0044	.9967
1.3262	.1548	.8136	.7894	.1333	.9673	.9546
1.9428	.1696	.7213	.6147	.3332	.9182	.8903
2.6302	.1770	.6362	.5340	.5075	.8754	.8280
3.4209	.1798	.5563	.4918	.6663	.8364	.7664
4.3867	.1806	.4787	.4727	.8126	.8005	.7026
5.2750	.1812	.4220	.4713	.9099	.7765	.6526
6.1687	.1825	.3755	.4785	.9795	.7595	.6090
7.1664	.1851	.3334	.4923	1.0309	.7468	.5667
8.0399	.1885	.3031	.5070	1.0588	.7400	.5342
8.8977	.1927	.2780	.5223	1.0748	.7361	.5058
9.8440	.1982	.2547	.5393	1.0831	.7340	.4777
10.9819	.2059	.2316	.5590	1.0845	.7337	.4478
12.1190	.2144	.2126	.5769	1.0806	.7346	.4215
13.2855	.2236	.1965	.5928	1.0750	.7360	.3975
14.5190	.2332	.1824	.6069	1.0699	.7373	.3749
15.8644	.2432	.1696	.6189	1.0673	.7379	.3530
17.3762	.2534	.1578	.6289	1.0683	.7377	.3313
19.1171	.2637	.1467	.6369	1.0738	.7363	.3094
21.1438	.2737	.1364	.6431	1.0839	.7338	.2873
23.5271	.2835	.1268	.6478	1.0981	.7303	.2650
26.1836	.2924	.1184	.6510	1.1146	.7263	.2439
29.0517	.3000	.1114	.6531	1.1319	.7220	.2246
32.1507	.3064	.1054	.6544	1.1496	.7177	.2069
35.5063	.3118	.1004	.6552	1.1671	.7134	.1907
39.1452	.3162	.0960	.6557	1.1842	.7092	.1757
43.0925	.3199	.0923	.6559	1.2004	.7052	.1619
47.3721	.3228	.0891	.6562	1.2153	.7016	.1492
52.0081	.3252	.0864	.6565	1.2287	.6983	.1375
57.0277	.3273	.0840	.6568	1.2406	.6953	.1268
62.4632	.3289	.0820	.6573	1.2511	.6928	.1169
68.3537	.3303	.0803	.6579	1.2603	.6905	.1078
74.7444	.3315	.0788	.6585	1.2683	.6886	.0994
82.3463	.3326	.0774	.6593	1.2758	.6867	.0909
89.9526	.3334	.0763	.6600	1.2817	.6852	.0838
98.2305	.3340	.0753	.6608	1.2869	.6840	.0772
107.2440	.3346	.0744	.6616	1.2914	.6829	.0712
117.0607	.3350	.0737	.6623	1.2953	.6819	.0655
127.7531	.3354	.0731	.6631	1.2987	.6811	.0603
139.3980	.3357	.0725	.6639	1.3016	.6804	.0556
152.0778	.3359	.0721	.6646	1.3039	.6798	.0511
165.8802	.3361	.0717	.6654	1.3059	.6793	.0471
180.8993	.3364	.0713	.6661	1.3074	.6789	.0433
200.3551	.3366	.0710	.6670	1.3088	.6786	.0392

MACH NO = 15.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0347	1.0391
.9663	.1401	.8692	1.0372	-.0179	1.0044	.9966
1.3767	.1550	.7986	.7664	.1526	.9625	.9490
1.9959	.1677	.7075	.6031	.3506	.9139	.8851
2.7685	.1736	.6144	.5193	.5434	.8666	.8166
3.6496	.1744	.5297	.4782	.7143	.8246	.7503
4.5834	.1735	.4586	.4630	.8502	.7912	.6908
5.5263	.1730	.4014	.4631	.9491	.7669	.6397
6.4538	.1724	.3559	.4718	1.0163	.7504	.5962
7.3526	.1750	.3198	.4851	1.0590	.7399	.5594
8.2156	.1777	.2908	.5003	1.0841	.7338	.5281
8.9499	.1809	.2698	.5142	1.0959	.7309	.5041
9.7405	.1949	.2502	.5296	1.1019	.7296	.4806
10.8277	.1918	.2275	.5507	1.0985	.7302	.4516
11.8750	.1996	.2094	.5700	1.0898	.7324	.4268
12.8350	.2075	.1955	.5860	1.0798	.7348	.4064
13.9022	.2164	.1823	.6012	1.0695	.7374	.3858
14.9482	.2250	.1713	.6135	1.0620	.7392	.3676
16.1974	.2346	.1602	.6248	1.0575	.7403	.3480
17.6216	.2444	.1495	.6340	1.0579	.7402	.3280
19.1125	.2532	.1402	.6406	1.0628	.7390	.3095
21.0040	.2628	.1305	.6462	1.0727	.7366	.2887
23.1086	.2718	.1218	.6503	1.0857	.7334	.2687
25.8091	.2812	.1130	.6534	1.1032	.7291	.2467
28.8749	.2895	.1053	.6554	1.1230	.7242	.2257
31.9436	.2959	.0993	.6563	1.1422	.7195	.2080
35.5675	.3015	.0936	.6565	1.1630	.7144	.1904
39.2187	.3056	.0892	.6566	1.1815	.7099	.1754
43.5302	.3093	.0851	.6566	1.1999	.7053	.1605
48.2318	.3123	.0817	.6567	1.2163	.7013	.1469
52.9411	.3147	.0789	.6569	1.2297	.6980	.1354
58.4798	.3167	.0764	.6573	1.2423	.6949	.1240
64.5240	.3185	.0743	.6579	1.2533	.6922	.1135
70.6054	.3198	.0726	.6585	1.2620	.6901	.1047
77.8006	.3211	.0710	.6592	1.2702	.6881	.0958
85.0616	.3220	.0697	.6599	1.2768	.6865	.0883
93.6671	.3229	.0685	.6607	1.2828	.6850	.0807
103.1172	.3237	.0675	.6616	1.2880	.6837	.0738
112.6618	.3243	.0666	.6624	1.2920	.6827	.0679
123.9761	.3249	.0659	.6633	1.2956	.6818	.0621
135.4004	.3254	.0653	.6642	1.2983	.6812	.0571
148.9352	.3259	.0648	.6651	1.3007	.6806	.0522
163.7814	.3264	.0643	.6660	1.3026	.6801	.0476
178.7504	.3269	.0640	.6669	1.3039	.6798	.0438
200.7722	.3276	.0636	.6680	1.3052	.6795	.0392

MACH NO = 20.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1285	.9279	1.3710	-.1405	1.0345	1.0388
1.0099	.1417	.8587	.9950	.0025	.9994	.9913
1.4332	.1560	.7871	.7435	.1733	.9574	.9420
2.0643	.1675	.6960	.5914	.3706	.9090	.8786
2.8420	.1722	.6042	.5125	.5615	.8621	.8106
3.7183	.1723	.5215	.4739	.7293	.8209	.7456
4.6372	.1709	.4526	.4595	.8619	.7884	.6877
5.5575	.1699	.3973	.4595	.9584	.7646	.6381
6.4560	.1699	.3534	.4678	1.0242	.7485	.5962
7.3201	.1711	.3186	.4804	1.0662	.7382	.5607
8.1429	.1732	.2907	.4949	1.0912	.7320	.5306
8.9219	.1761	.2681	.5099	1.1042	.7288	.5050
9.7382	.1801	.2478	.5262	1.1089	.7277	.4807
10.7346	.1863	.2268	.5464	1.1054	.7286	.4540
11.6811	.1933	.2101	.5649	1.0958	.7309	.4312
12.6769	.2014	.1952	.5829	1.0831	.7340	.4096
13.6109	.2095	.1833	.5976	1.0715	.7369	.3912
14.6660	.2185	.1718	.6113	1.0613	.7394	.3724
15.7357	.2271	.1617	.6222	1.0551	.7409	.3550
17.0361	.2365	.1513	.6320	1.0532	.7414	.3360
18.3921	.2451	.1422	.6390	1.0562	.7406	.3182
19.9554	.2536	.1333	.6446	1.0634	.7389	.2999
21.9814	.2631	.1239	.6493	1.0754	.7359	.2790
24.2437	.2719	.1156	.6528	1.0899	.7323	.2590
27.0700	.2809	.1073	.6553	1.1085	.7278	.2376
30.1086	.2882	.1004	.6565	1.1285	.7229	.2183
33.6910	.2945	.0940	.6569	1.1512	.7173	.1991
37.3424	.2992	.0890	.6568	1.1717	.7123	.1828
41.3306	.3030	.0847	.6566	1.1907	.7076	.1678
46.0163	.3064	.0807	.6566	1.2089	.7031	.1530
50.7498	.3090	.0776	.6568	1.2237	.6995	.1405
56.2988	.3113	.0748	.6572	1.2375	.6961	.1282
61.9222	.3131	.0726	.6577	1.2487	.6934	.1178
68.0576	.3146	.0707	.6593	1.2584	.6910	.1082
75.3064	.3160	.0689	.6590	1.2673	.6888	.0987
82.6980	.3171	.0674	.6598	1.2744	.6870	.0906
91.4437	.3181	.0661	.6607	1.2809	.6854	.0826
100.3657	.3189	.0651	.6615	1.2860	.6842	.0757
110.9236	.3197	.0641	.6625	1.2905	.6831	.0689
121.6946	.3204	.0633	.6634	1.2939	.6822	.0632
133.4847	.3211	.0627	.6644	1.2967	.6816	.0579
147.4261	.3217	.0621	.6654	1.2991	.6810	.0527
161.6296	.3224	.0617	.6663	1.3008	.6806	.0482
178.4010	.3230	.0613	.6673	1.3021	.6802	.0439
201.0176	.3239	.0610	.6685	1.3033	.6800	.0391

MACH NO = 25.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0344	1.0386
1.0097	.1416	.8577	.9951	.0024	.9994	.9914
1.4313	.1557	.7864	.7440	.1727	.9576	.9430
2.0595	.1670	.6956	.5916	.3697	.9092	.8790
2.9256	.1717	.5944	.5063	.5805	.8574	.8039
3.8029	.1712	.5135	.4704	.7450	.8171	.7398
4.7153	.1696	.4464	.4575	.8739	.7854	.6832
5.6238	.1684	.3928	.4581	.9674	.7624	.6348
6.6027	.1683	.3460	.4677	1.0365	.7455	.5898
7.4436	.1695	.3130	.4804	1.0751	.7360	.5560
8.2402	.1715	.2866	.4947	1.0977	.7304	.5273
8.9910	.1743	.2652	.5093	1.1093	.7276	.5028
9.7745	.1781	.2459	.5252	1.1130	.7267	.4796
10.7972	.1844	.2245	.5463	1.1081	.7279	.4524
11.6937	.1911	.2088	.5643	1.0978	.7304	.4309
12.6324	.1988	.1947	.5817	1.0846	.7337	.4105
13.5752	.2070	.1827	.5971	1.0717	.7368	.3919
14.5597	.2155	.1718	.6105	1.0611	.7394	.3742
15.6318	.2243	.1616	.6219	1.0539	.7412	.3566
16.8391	.2332	.1517	.6314	1.0513	.7418	.3387
18.1886	.2419	.1424	.6387	1.0538	.7412	.3207
19.6285	.2500	.1340	.6441	1.0602	.7396	.3035
21.4789	.2589	.1250	.6488	1.0710	.7370	.2839
23.6793	.2679	.1164	.6525	1.0849	.7336	.2637
26.1884	.2764	.1087	.6551	1.1013	.7295	.2439
29.2015	.2843	.1013	.6567	1.1214	.7246	.2237
32.6222	.2908	.0947	.6570	1.1440	.7191	.2045
36.1069	.2956	.0895	.6569	1.1648	.7140	.1880
40.2194	.2998	.0846	.6567	1.1857	.7088	.1717
44.7152	.3033	.0806	.6566	1.2043	.7043	.1569
49.6117	.3061	.0772	.6568	1.2204	.7003	.1433
54.9468	.3085	.0743	.6571	1.2345	.6969	.1310
60.7789	.3104	.0718	.6576	1.2466	.6939	.1198
67.1757	.3121	.0697	.6582	1.2571	.6913	.1095
73.6863	.3134	.0680	.6589	1.2654	.6893	.1007
81.3753	.3146	.0664	.6597	1.2731	.6874	.0919
89.8388	.3156	.0651	.6606	1.2796	.6858	.0839
99.1553	.3166	.0639	.6615	1.2850	.6845	.0766
109.4118	.3174	.0629	.6625	1.2894	.6834	.0698
120.7028	.3182	.0621	.6635	1.2931	.6825	.0637
133.1293	.3189	.0614	.6645	1.2960	.6817	.0580
145.7780	.3196	.0609	.6655	1.2981	.6812	.0532
160.7006	.3204	.0605	.6665	1.2998	.6808	.0485
177.0894	.3211	.0601	.6675	1.3011	.6805	.0442
200.5231	.3221	.0597	.6687	1.3022	.6802	.0392

MACH NO = 30.00 CONE ANGLE = 7.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0344	1.0385
1.0096	.1415	.8571	.9952	.0024	.9994	.9914
1.4302	.1555	.7859	.7444	.1724	.9577	.9431
2.0567	.1667	.6954	.5918	.3691	.9094	.8793
2.9201	.1712	.5944	.5062	.5799	.8576	.8043
3.7938	.1707	.5137	.4700	.7444	.8172	.7404
4.7016	.1689	.4467	.4567	.8736	.7855	.6840
5.6044	.1676	.3932	.4569	.9675	.7624	.6358
6.5759	.1674	.3466	.4662	1.0372	.7453	.5910
7.4088	.1684	.3137	.4787	1.0761	.7357	.5573
8.1966	.1703	.2873	.4928	1.0992	.7301	.5288
8.9375	.1729	.2660	.5072	1.1111	.7271	.5045
9.7086	.1765	.2468	.5229	1.1152	.7261	.4815
10.7114	.1825	.2256	.5438	1.1106	.7273	.4546
11.6516	.1895	.2089	.5630	1.0995	.7300	.4319
12.5601	.1970	.1951	.5803	1.0861	.7333	.4120
13.4675	.2049	.1833	.5955	1.0730	.7365	.3940
14.4782	.2137	.1720	.6097	1.0612	.7394	.3756
15.5025	.2222	.1622	.6210	1.0536	.7413	.3587
16.6513	.2309	.1526	.6305	1.0503	.7421	.3414
17.9280	.2393	.1435	.6378	1.0520	.7417	.3240
19.3912	.2476	.1347	.6436	1.0582	.7401	.3062
21.1353	.2562	.1260	.6483	1.0682	.7377	.2874
23.1975	.2650	.1176	.6521	1.0811	.7345	.2679
25.6937	.2738	.1095	.6550	1.0973	.7305	.2476
28.4791	.2815	.1023	.6566	1.1159	.7260	.2282
31.7838	.2882	.0956	.6571	1.1383	.7205	.2089
35.4253	.2935	.0898	.6569	1.1609	.7149	.1910
39.4301	.2978	.0849	.6567	1.1820	.7097	.1746
43.8064	.3013	.0807	.6566	1.2009	.7051	.1597
48.5677	.3043	.0772	.6567	1.2173	.7011	.1460
54.1357	.3068	.0740	.6570	1.2326	.6973	.1328
59.8279	.3088	.0715	.6575	1.2448	.6943	.1215
66.0674	.3105	.0693	.6581	1.2554	.6917	.1112
72.9227	.3119	.0675	.6588	1.2645	.6895	.1016
80.4611	.3132	.0659	.6596	1.2722	.6876	.0929
88.7516	.3142	.0645	.6605	1.2786	.6860	.0849
98.5546	.3153	.0633	.6615	1.2844	.6846	.0770
108.6522	.3161	.0623	.6625	1.2888	.6835	.0703
119.7588	.3169	.0615	.6635	1.2925	.6826	.0641
131.9721	.3177	.0608	.6645	1.2953	.6819	.0585
145.3942	.3185	.0602	.6656	1.2976	.6814	.0534
160.1334	.3193	.0598	.6666	1.2992	.6810	.0487
176.3048	.3201	.0594	.6676	1.3005	.6806	.0444
200.7656	.3211	.0591	.6689	1.3016	.6804	.0392

MACH NO = 3.50 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0469	1.0542
.8518	.1395	.9807	1.1740	-.0749	1.0211	1.0112
1.0607	.1534	.9361	.9561	.0229	.9936	.9820
1.3874	.1728	.8734	.7791	.1440	.9595	.9396
1.8896	.1961	.7910	.6568	.2857	.9197	.8812
2.4312	.2136	.7171	.5969	.4047	.8863	.8258
3.0744	.2277	.6447	.5623	.5170	.8547	.7684
3.8096	.2391	.5779	.5453	.6166	.8267	.7119
4.7786	.2505	.5091	.5399	.7134	.7995	.6490
5.7042	.2594	.4582	.5431	.7799	.7808	.5985
6.7100	.2679	.4148	.5502	.8327	.7660	.5518
7.7879	.2758	.3793	.5590	.8744	.7542	.5092
9.1342	.2843	.3433	.5699	.9124	.7435	.4645
10.5774	.2923	.3148	.5804	.9420	.7352	.4245
12.1216	.2996	.2917	.5901	.9655	.7286	.3887
13.7722	.3063	.2727	.5989	.9848	.7232	.3565
15.5364	.3123	.2572	.6066	1.0011	.7186	.3276
17.4244	.3177	.2444	.6133	1.0154	.7146	.3014
19.4498	.3225	.2337	.6192	1.0280	.7110	.2776
21.6279	.3267	.2248	.6243	1.0394	.7079	.2558
23.9751	.3304	.2173	.6288	1.0497	.7050	.2359
26.5078	.3337	.2111	.6329	1.0591	.7023	.2176
29.2432	.3365	.2058	.6364	1.0676	.6999	.2008
32.1991	.3390	.2014	.6397	1.0754	.6977	.1854
35.3945	.3412	.1976	.6426	1.0824	.6958	.1711
38.7656	.3434	.1940	.6456	1.0897	.6937	.1562
43.1434	.3450	.1914	.6480	1.0953	.6921	.1442
47.2284	.3467	.1892	.6502	1.1005	.6907	.1332
51.6453	.3476	.1874	.6522	1.1052	.6894	.1230
56.4212	.3485	.1858	.6540	1.1094	.6882	.1137
61.5856	.3493	.1845	.6557	1.1133	.6871	.1050
67.1700	.3499	.1833	.6572	1.1167	.6861	.0970
73.2093	.3504	.1824	.6586	1.1199	.6852	.0896
79.7412	.3508	.1815	.6599	1.1227	.6844	.0828
86.8067	.3511	.1808	.6612	1.1253	.6837	.0765
94.4504	.3513	.1802	.6623	1.1275	.6831	.0707
102.7208	.3515	.1797	.6634	1.1295	.6825	.0653
111.6707	.3516	.1792	.6644	1.1313	.6820	.0604
121.3572	.3516	.1789	.6654	1.1328	.6816	.0558
131.8420	.3516	.1785	.6663	1.1341	.6812	.0516
143.1922	.3516	.1782	.6671	1.1352	.6809	.0476
157.3167	.3515	.1779	.6680	1.1363	.6806	.0435
170.7727	.3514	.1777	.6688	1.1371	.6804	.0402
185.3421	.3513	.1775	.6695	1.1378	.6802	.0372
201.1177	.3512	.1773	.6702	1.1384	.6800	.0343

MACH NO = 5.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0430	1.0457
.8763	.1386	.9288	1.1415	-.0624	1.0176	1.0076
1.1594	.1536	.8706	.8855	.0643	.9819	.9688
1.5926	.1713	.7925	.7073	.2133	.9401	.9148
2.1318	.1859	.7106	.6098	.3559	.9000	.8555
2.7734	.1963	.6302	.5564	.4885	.8627	.7942
3.5037	.2038	.5564	.5302	.6044	.8301	.7344
4.3060	.2100	.4915	.5216	.6986	.8036	.6782
5.1652	.2161	.4363	.5240	.7706	.7834	.6269
6.0725	.2226	.3900	.5328	.8235	.7685	.5805
7.0261	.2296	.3512	.5447	.8615	.7578	.5386
7.8831	.2361	.3228	.5559	.8854	.7511	.5058
8.9358	.2440	.2945	.5691	.9058	.7454	.4706
10.3866	.2548	.2642	.5853	.9247	.7401	.4294
11.7872	.2646	.2418	.5982	.9375	.7365	.3959
13.3204	.2743	.2228	.6095	.9487	.7333	.3648
15.2294	.2851	.2049	.6202	.9610	.7299	.3323
17.0425	.2938	.1919	.6277	.9718	.7268	.3063
18.9743	.3017	.1811	.6337	.9829	.7237	.2828
21.3119	.3095	.1710	.6390	.9955	.7202	.2588
23.5546	.3157	.1636	.6427	1.0069	.7170	.2393
25.9696	.3211	.1573	.6456	1.0182	.7138	.2213
28.9124	.3263	.1514	.6483	1.0306	.7103	.2027
31.7463	.3303	.1470	.6503	1.0412	.7073	.1876
34.8028	.3337	.1432	.6520	1.0513	.7045	.1736
38.5297	.3369	.1397	.6536	1.0619	.7015	.1591
42.1204	.3393	.1370	.6548	1.0708	.6990	.1473
45.9940	.3412	.1347	.6559	1.0790	.6967	.1364
50.7178	.3430	.1325	.6570	1.0875	.6943	.1250
55.2691	.3443	.1309	.6580	1.0943	.6924	.1158
60.1799	.3453	.1294	.6589	1.1005	.6907	.1072
66.1706	.3462	.1281	.6599	1.1066	.6890	.0983
71.9455	.3469	.1270	.6608	1.1113	.6876	.0911
78.1803	.3473	.1261	.6616	1.1155	.6864	.0843
85.7922	.3478	.1252	.6626	1.1196	.6853	.0774
93.1359	.3481	.1246	.6635	1.1227	.6844	.0716
101.0708	.3483	.1240	.6643	1.1255	.6836	.0663
110.7658	.3484	.1234	.6652	1.1281	.6829	.0608
120.1260	.3485	.1229	.6660	1.1302	.6823	.0563
130.2454	.3485	.1226	.6667	1.1319	.6818	.0522
142.6154	.3485	.1222	.6676	1.1336	.6814	.0478
154.5620	.3484	.1219	.6683	1.1348	.6810	.0443
167.4799	.3484	.1216	.6689	1.1359	.6807	.0410
183.2715	.3483	.1214	.6697	1.1369	.6804	.0376
200.5128	.3482	.1212	.6704	1.1376	.6802	.0344

MACH NO = 10.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0402	1.0401
.9659	.1405	.8737	1.0389	-.0187	1.0053	.9950
1.3039	.1535	.8078	.8043	.1212	.9659	.9501
1.8694	.1668	.7136	.6361	.3004	.9155	.8834
2.4880	.1735	.6290	.5570	.4521	.8729	.8204
3.1911	.1762	.5508	.5148	.5875	.8349	.7589
4.0413	.1773	.4756	.4950	.7100	.8004	.6958
4.8174	.1782	.4209	.4926	.7904	.7778	.6467
5.5940	.1799	.3763	.4985	.8472	.7619	.6041
6.4573	.1829	.3358	.5107	.8891	.7501	.5628
7.2105	.1864	.3068	.5239	.9117	.7437	.5312
7.9483	.1907	.2827	.5377	.9249	.7400	.5034
8.7603	.1961	.2604	.5531	.9319	.7380	.4761
9.7342	.2035	.2381	.5708	.9336	.7376	.4470
10.7930	.2124	.2184	.5884	.9309	.7383	.4191
11.7859	.2210	.2031	.6027	.9271	.7394	.3959
12.9285	.2307	.1886	.6162	.9235	.7404	.3723
14.0668	.2399	.1767	.6268	.9222	.7408	.3513
15.4576	.2501	.1647	.6364	.9240	.7403	.3288
16.9208	.2595	.1544	.6434	.9290	.7389	.3079
18.7768	.2696	.1440	.6493	.9384	.7362	.2850
20.7820	.2786	.1351	.6534	.9504	.7329	.2639
23.3423	.2878	.1263	.6565	.9662	.7284	.2410
25.9375	.2951	.1195	.6582	.9818	.7240	.2215
29.0151	.3017	.1132	.6594	.9989	.7192	.2021
32.0873	.3067	.1083	.6599	1.0144	.7149	.1859
35.7426	.3112	.1038	.6602	1.0305	.7103	.1697
39.3968	.3145	.1003	.6604	1.0442	.7065	.1561
43.7427	.3175	.0971	.6606	1.0578	.7027	.1425
48.0825	.3197	.0946	.6609	1.0689	.6996	.1311
53.2400	.3216	.0923	.6613	1.0795	.6966	.1197
58.3920	.3231	.0905	.6618	1.0880	.6942	.1102
64.5234	.3244	.0888	.6624	1.0959	.6920	.1006
70.6608	.3255	.0875	.6631	1.1022	.6902	.0926
77.9799	.3264	.0862	.6638	1.1081	.6885	.0845
85.3186	.3271	.0852	.6646	1.1127	.6872	.0778
93.3184	.3277	.0844	.6653	1.1167	.6861	.0715
102.8717	.3282	.0836	.6661	1.1204	.6851	.0652
112.4567	.3286	.0830	.6669	1.1232	.6843	.0600
123.9013	.3290	.0824	.6678	1.1258	.6835	.0547
135.3798	.3293	.0820	.6685	1.1278	.6830	.0503
149.0775	.3296	.0816	.6693	1.1295	.6825	.0458
162.8063	.3299	.0813	.6701	1.1307	.6822	.0421
179.1771	.3302	.0811	.6709	1.1317	.6819	.0384
200.2883	.3306	.0808	.6718	1.1326	.6816	.0345

MACH NO = 15.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0397	1.0391
.9659	.1399	.8668	1.0388	-.0186	1.0052	.9950
1.3507	.1535	.7925	.7824	.1388	.9610	.9442
1.9171	.1649	.6999	.6250	.3157	.9113	.8782
2.6108	.1702	.6075	.5427	.4829	.8643	.8089
3.3921	.1712	.5247	.5017	.6279	.8235	.7429
4.1209	.1709	.4628	.4867	.7301	.7948	.6904
4.9455	.1708	.4062	.4846	.8147	.7710	.6393
5.7553	.1717	.3612	.4916	.8719	.7549	.5959
6.5397	.1737	.3255	.5033	.9082	.7447	.5592
7.2930	.1766	.2969	.5171	.9295	.7387	.5279
7.9350	.1798	.2761	.5299	.9399	.7358	.5039
8.6274	.1840	.2567	.5441	.9447	.7345	.4804
9.5812	.1909	.2342	.5635	.9437	.7347	.4513
10.5010	.1986	.2163	.5812	.9376	.7365	.4264
11.4143	.2068	.2014	.5971	.9297	.7387	.4043
12.3502	.2154	.1886	.6109	.9223	.7407	.3839
13.3421	.2242	.1771	.6228	.9169	.7423	.3644
14.4313	.2332	.1664	.6328	.9145	.7430	.3451
15.7614	.2430	.1555	.6415	.9159	.7426	.3242
17.1670	.2520	.1461	.6478	.9212	.7411	.3047
18.8107	.2609	.1370	.6526	.9302	.7385	.2847
20.7886	.2699	.1283	.6563	.9425	.7351	.2638
23.1452	.2785	.1201	.6589	.9577	.7308	.2426
25.9125	.2865	.1127	.6604	.9753	.7259	.2217
28.9796	.2931	.1063	.6611	.9937	.7207	.2023
32.3415	.2983	.1009	.6612	1.0119	.7156	.1847
36.0301	.3026	.0964	.6612	1.0289	.7108	.1685
40.0717	.3061	.0926	.6612	1.0443	.7065	.1538
44.4905	.3089	.0893	.6613	1.0579	.7026	.1404
49.3185	.3113	.0866	.6616	1.0698	.6993	.1282
54.5983	.3132	.0843	.6620	1.0802	.6964	.1171
60.3830	.3148	.0823	.6625	1.0891	.6939	.1069
66.7322	.3161	.0807	.6632	1.0967	.6917	.0976
73.7100	.3173	.0793	.6639	1.1032	.6899	.0891
81.3836	.3182	.0780	.6647	1.1088	.6883	.0813
89.8242	.3191	.0770	.6655	1.1134	.6871	.0741
99.1086	.3198	.0761	.6664	1.1172	.6860	.0676
109.3196	.3204	.0754	.6673	1.1204	.6851	.0616
120.5472	.3211	.0748	.6682	1.1229	.6844	.0561
132.8881	.3216	.0743	.6691	1.1250	.6838	.0512
146.4465	.3222	.0739	.6700	1.1266	.6833	.0466
161.3349	.3227	.0736	.6708	1.1278	.6830	.0425
177.6746	.3233	.0733	.6717	1.1288	.6827	.0387
201.4527	.3240	.0730	.6727	1.1297	.6825	.0343

MACH NO = 20.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	XCP/D	XVCP/LV	RN/RB
.7294	.1285	.0270	1.3710	-.1405	1.0395	1.0388
.8659	.1706	.0642	1.0348	-.1187	1.0052	.9950
1.7477	.1509	.7006	.7835	.1378	.9613	.9446
1.9083	.1678	.6987	.6256	.3140	.9117	.8791
2.5970	.1688	.6071	.5425	.4809	.8648	.8105
3.3630	.1693	.5251	.5034	.6261	.8240	.7451
4.1690	.1685	.4566	.4834	.7400	.7920	.6871
4.9757	.1691	.4019	.4815	.8224	.7688	.6375
5.7618	.1686	.3586	.4882	.8782	.7532	.5956
6.5191	.1712	.3230	.4934	.9137	.7472	.5601
7.2395	.1726	.2963	.5127	.9348	.7372	.5300
7.9247	.1758	.2770	.5265	.9460	.7341	.5043
8.5741	.1795	.2555	.5402	.9503	.7329	.4821
9.1869	.1862	.2328	.5514	.9483	.7374	.4529
10.4327	.1933	.2151	.5738	.9407	.7356	.4282
11.7209	.2012	.2013	.5952	.9312	.7382	.4065
12.2217	.2113	.1977	.6035	.9224	.7417	.3866
13.0987	.2194	.1772	.6210	.9159	.7425	.3690
14.1267	.2272	.1658	.6314	.9119	.7437	.3503
15.0796	.2362	.1568	.6410	.9117	.7437	.3315
15.5774	.2449	.1474	.6456	.9156	.7426	.3126
16.1745	.2575	.1394	.6517	.9232	.7415	.2933
17.2362	.2618	.1313	.6555	.9332	.7377	.2745
21.8321	.2795	.1220	.6585	.9467	.7379	.2540
24.0749	.2787	.1145	.6605	.9626	.7294	.2336
27.2391	.2861	.1073	.6615	.9816	.7241	.2129
32.5696	.2922	.1011	.6616	1.0014	.7185	.1936
37.9688	.2967	.0963	.6614	1.0190	.7136	.1772
37.9877	.3007	.0910	.6613	1.0363	.7087	.1611
42.7997	.3029	.0882	.6613	1.0515	.7045	.1465
47.2352	.3065	.0851	.6616	1.0647	.7007	.1332
52.1451	.3086	.0826	.6619	1.0753	.6978	.1220
57.0414	.3104	.0814	.6624	1.0852	.6950	.1110
64.7316	.3120	.0795	.6631	1.0936	.6926	.1009
71.3868	.3132	.0780	.6638	1.1007	.6906	.0917
79.1836	.3143	.0766	.6646	1.1067	.6889	.0834
87.1471	.3152	.0745	.6655	1.1113	.6876	.0762
96.5918	.3161	.0736	.6664	1.1155	.6865	.0692
107.0244	.3169	.0727	.6673	1.1188	.6855	.0629
118.5453	.3177	.0721	.6683	1.1215	.6848	.0570
131.2627	.3184	.0715	.6693	1.1236	.6842	.0518
144.2445	.3190	.0711	.6702	1.1251	.6838	.0473
159.6378	.3197	.0708	.6711	1.1264	.6834	.0429
176.5315	.3204	.0706	.6720	1.1274	.6831	.0389
200.8317	.3213	.0702	.6731	1.1283	.6829	.0344

MACH NO = 25.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0394	1.0386
1.0072	.1413	.8547	.9991	.0005	.9999	.9893
1.4011	.1541	.7800	.7610	.1570	.9559	.9379
1.9744	.1642	.6880	.6138	.3324	.9066	.8720
2.6678	.1683	.5974	.5360	.4975	.8602	.8037
3.4387	.1684	.5169	.4968	.6396	.8202	.7393
4.2392	.1674	.4504	.4814	.7503	.7891	.6826
5.0352	.1668	.3972	.4801	.8300	.7667	.6341
5.8085	.1672	.3551	.4870	.8838	.7516	.5933
6.5496	.1686	.3215	.4982	.9181	.7419	.5587
7.2540	.1709	.2947	.5113	.9384	.7362	.5294
7.9203	.1739	.2729	.5249	.9492	.7332	.5044
8.6183	.1778	.2532	.5397	.9534	.7320	.4807
9.5334	.1843	.2315	.5595	.9505	.7328	.4527
10.3994	.1915	.2144	.5776	.9424	.7351	.4290
11.2438	.1992	.2003	.5937	.9326	.7379	.4082
12.0939	.2073	.1883	.6078	.9232	.7405	.3893
13.0458	.2161	.1767	.6208	.9153	.7427	.3700
14.0122	.2246	.1667	.6310	.9108	.7440	.3523
15.0901	.2331	.1572	.6394	.9100	.7442	.3344
16.2879	.2414	.1483	.6459	.9131	.7434	.3166
17.6630	.2497	.1396	.6510	.9197	.7415	.2984
19.4260	.2587	.1306	.6553	.9302	.7385	.2778
21.3353	.2669	.1227	.6583	.9424	.7351	.2585
23.5195	.2747	.1155	.6604	.9566	.7311	.2395
26.1541	.2820	.1086	.6616	.9737	.7263	.2200
29.5469	.2888	.1017	.6618	.9950	.7203	.1991
33.0557	.2938	.0963	.6616	1.0143	.7149	.1813
36.9193	.2979	.0917	.6614	1.0319	.7100	.1651
41.1549	.3012	.0879	.6613	1.0474	.7056	.1503
45.7885	.3039	.0847	.6615	1.0609	.7018	.1369
51.2431	.3064	.0818	.6619	1.0734	.6983	.1239
56.8525	.3082	.0795	.6624	1.0833	.6955	.1129
63.0289	.3098	.0776	.6630	1.0918	.6931	.1028
69.8394	.3111	.0760	.6637	1.0990	.6911	.0936
77.3524	.3123	.0746	.6645	1.1051	.6894	.0852
86.2642	.3134	.0733	.6655	1.1105	.6879	.0770
95.4712	.3143	.0723	.6664	1.1146	.6867	.0700
105.6265	.3151	.0715	.6674	1.1179	.6858	.0636
116.8253	.3159	.0708	.6683	1.1206	.6850	.0578
130.0977	.3167	.0703	.6694	1.1228	.6844	.0522
143.7902	.3175	.0698	.6704	1.1244	.6840	.0474
158.8634	.3182	.0695	.6713	1.1256	.6836	.0431
175.4457	.3190	.0692	.6722	1.1266	.6833	.0392
200.6163	.3200	.0689	.6733	1.1276	.6831	.0344

MACH NO = 30.00 CONE ANGLE = 8.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0394	1.0385
1.0070	.1412	.8541	.9992	.0004	.9999	.9893
1.4000	.1539	.7796	.7613	.1567	.9560	.9380
1.9718	.1639	.6877	.6140	.3320	.9067	.8723
2.6631	.1679	.5973	.5359	.4970	.8603	.8041
3.4310	.1679	.5171	.4965	.6392	.8203	.7399
4.3163	.1667	.4442	.4800	.7604	.7863	.6776
5.1057	.1660	.3924	.4796	.8372	.7647	.6302
5.8704	.1664	.3514	.4869	.8889	.7501	.5902
6.6012	.1679	.3188	.4982	.9217	.7409	.5565
7.2943	.1701	.2926	.5112	.9411	.7355	.5279
7.9486	.1730	.2714	.5246	.9512	.7326	.5034
8.6327	.1769	.2522	.5392	.9550	.7316	.4802
9.5276	.1831	.2310	.5588	.9518	.7325	.4528
10.4315	.1907	.2132	.5779	.9428	.7350	.4282
11.2518	.1982	.1996	.5937	.9327	.7378	.4081
12.0759	.2061	.1879	.6076	.9232	.7405	.3896
12.9964	.2147	.1767	.6204	.9151	.7428	.3709
13.9278	.2230	.1669	.6305	.9103	.7441	.3538
15.0406	.2318	.1570	.6394	.9092	.7445	.3352
16.1940	.2399	.1483	.6458	.9120	.7437	.3179
17.6152	.2485	.1393	.6511	.9188	.7418	.2990
19.1951	.2566	.1311	.6550	.9281	.7391	.2807
21.1356	.2652	.1229	.6583	.9405	.7356	.2604
23.1966	.2727	.1159	.6604	.9539	.7319	.2422
25.8543	.2804	.1087	.6617	.9713	.7270	.2221
28.9113	.2867	.1022	.6619	.9909	.7215	.2027
32.3220	.2919	.0967	.6617	1.0104	.7160	.1848
36.3595	.2963	.0917	.6614	1.0295	.7106	.1672
40.5018	.2997	.0878	.6613	1.0452	.7062	.1524
45.3675	.3027	.0843	.6615	1.0598	.7021	.1380
50.3509	.3050	.0815	.6618	1.0715	.6988	.1259
56.2297	.3070	.0791	.6623	1.0822	.6958	.1140
62.2870	.3086	.0771	.6630	1.0908	.6934	.1039
69.4635	.3101	.0753	.6637	1.0985	.6912	.0941
76.8721	.3112	.0739	.6645	1.1046	.6895	.0857
85.6525	.3123	.0727	.6655	1.1099	.6880	.0775
94.7160	.3133	.0717	.6664	1.1140	.6869	.0705
104.7052	.3141	.0709	.6674	1.1174	.6859	.0642
116.5398	.3150	.0701	.6684	1.1202	.6851	.0580
128.7465	.3158	.0696	.6694	1.1223	.6846	.0527
143.1918	.3166	.0691	.6704	1.1240	.6841	.0476
158.0707	.3174	.0687	.6714	1.1252	.6837	.0433
175.6531	.3182	.0685	.6723	1.1262	.6835	.0391
200.6185	.3192	.0682	.6734	1.1271	.6832	.0344

MACH NO = 3.50 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0529	1.0542
.8518	.1396	.9808	1.1742	-.0750	1.0238	1.0111
1.0601	.1534	.9318	.9595	.0210	.9933	.9785
1.3835	.1723	.8640	.7870	.1373	.9565	.9318
1.7975	.1914	.7899	.6813	.2515	.9203	.8781
2.3027	.2082	.7144	.6186	.3603	.8859	.8205
2.9913	.2235	.6316	.5788	.4745	.8497	.7531
3.6613	.2340	.5679	.5628	.5582	.8232	.6974
4.4010	.2432	.5119	.5575	.6277	.8012	.6447
5.2082	.2517	.4634	.5591	.6838	.7834	.5956
6.0839	.25[illegible]	.4220	.5648	.7284	.7693	.5502
7.0328	.2677	.3866	.5724	.7640	.7580	.5081
8.2380	.2764	.3521	.5822	.7971	.7475	.4632
9.5299	.2845	.3242	.5918	.8229	.7393	.4231
10.9994	.2919	.3016	.6008	.8434	.7328	.3873
12.5996	.2996	.2809	.6099	.8624	.7268	.3509
14.1860	.3056	.2662	.6168	.8764	.7224	.3225
15.8836	.3110	.2541	.6229	.8888	.7185	.2968
17.9763	.3164	.2429	.6288	.9014	.7145	.2702
19.9567	.3205	.2347	.6333	.9114	.7113	.2491
22.4096	.3246	.2271	.6378	.9217	.7080	.2271
24.7384	.3277	.2215	.6412	.9299	.7054	.2095
27.2538	.3304	.2168	.6443	.9375	.7030	.1934
30.3782	.3330	.2124	.6474	.9453	.7006	.1765
33.3499	.3349	.2091	.6499	.9515	.6986	.1630
36.5632	.3366	.2064	.6521	.9571	.6968	.1505
40.5567	.3382	.2038	.6544	.9629	.6950	.1374
44.3564	.3394	.2019	.6563	.9674	.6935	.1269
48.4655	.3404	.2003	.6580	.9716	.6922	.1172
53.5734	.3413	.1987	.6598	.9758	.6909	.1071
58.4342	.3419	.1976	.6612	.9791	.6898	.0989
64.4775	.3424	.1965	.6628	.9825	.6888	.0904
70.2297	.3428	.1957	.6640	.9850	.6880	.0835
76.4531	.3431	.1950	.6652	.9873	.6872	.0771
84.1935	.3433	.1943	.6664	.9896	.6865	.0705
91.5642	.3434	.1938	.6675	.9913	.6860	.0651
99.5419	.3435	.1934	.6685	.9928	.6855	.0602
109.4683	.3436	.1930	.6695	.9943	.6850	.0550
118.9238	.3436	.1926	.6704	.9954	.6847	.0508
129.1611	.3435	.1924	.6712	.9963	.6844	.0469
141.9021	.3435	.1921	.6721	.9972	.6841	.0429
154.0412	.3434	.1919	.6728	.9978	.6839	.0396
169.1501	.3433	.1917	.6736	.9985	.6837	.0362
183.5458	.3433	.1915	.6743	.9989	.6836	.0334
201.4635	.3432	.1914	.6750	.9993	.6835	.0305

MACH NO = 5.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0484	1.0457
.8763	.1386	.9281	1.1417	-.0625	1.0198	1.0072
1.1436	.1528	.8660	.8962	.0576	.9818	.9653
1.5487	.1689	.7869	.7267	.1925	.9390	.9096
2.0376	.1822	.7057	.6317	.3188	.8990	.8498
2.6117	.1919	.6274	.5786	.4341	.8625	.7888
3.2587	.1990	.5560	.5513	.5336	.8310	.7298
3.9629	.2050	.4937	.5410	.6138	.8056	.6749
4.7106	.2109	.4409	.5415	.6748	.7862	.6249
5.4932	.2172	.3967	.5484	.7195	.7721	.5800
6.3083	.2239	.3597	.5584	.7516	.7619	.5396
7.0347	.2300	.3328	.5681	.7719	.7555	.5081
7.9188	.2375	.3059	.5797	.7893	.7500	.4743
9.1235	.2475	.2770	.5941	.8054	.7449	.4350
10.4231	.2578	.2533	.6070	.8175	.7410	.3992
11.8479	.2680	.2334	.6183	.8280	.7377	.3662
13.2529	.2770	.2183	.6269	.8372	.7348	.3386
15.0274	.2867	.2036	.6349	.8482	.7313	.3092
16.9595	.2955	.1916	.6411	.8596	.7277	.2825
19.0464	.3032	.1819	.6459	.8712	.7240	.2583
21.0491	.3091	.1747	.6492	.8817	.7207	.2388
23.4876	.3149	.1679	.6521	.8933	.7170	.2186
26.1420	.3199	.1623	.6543	.9046	.7134	.2002
29.0328	.3240	.1576	.6561	.9155	.7100	.1834
31.8189	.3271	.1541	.6575	.9246	.7071	.1697
35.2184	.3299	.1508	.6588	.9342	.7041	.1555
38.9234	.3322	.1481	.6599	.9430	.7013	.1425
42.9612	.3341	.1457	.6609	.9511	.6987	.1306
47.3619	.3355	.1438	.6619	.9583	.6965	.1197
51.6051	.3365	.1423	.6627	.9640	.6946	.1108
56.7847	.3375	.1409	.6637	.9696	.6929	.1015
62.4325	.3382	.1397	.6646	.9745	.6913	.0931
68.5925	.3387	.1387	.6656	.9788	.6900	.0853
74.5372	.3391	.1379	.6664	.9820	.6889	.0790
81.8007	.3394	.1371	.6674	.9851	.6879	.0724
89.7294	.3396	.1365	.6683	.9878	.6871	.0664
98.3863	.3398	.1359	.6692	.9901	.6864	.0608
106.7477	.3399	.1355	.6700	.9918	.6858	.0563
116.9717	.3399	.1351	.6708	.9935	.6853	.0516
128.1384	.3399	.1347	.6716	.9948	.6849	.0473
140.3349	.3399	.1345	.6724	.9960	.6845	.0433
152.1168	.3399	.1342	.6731	.9968	.6843	.0401
166.5224	.3398	.1340	.6738	.9975	.6840	.0367
182.2529	.3398	.1339	.6745	.9981	.6838	.0336
201.4310	.3397	.1337	.6752	.9985	.6837	.0305

MACH NO = 10.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0453	1.0401
.9651	.1405	.8713	1.0412	-.0198	1.0063	.9931
1.2878	.1526	.8019	.8169	.1114	.9647	.9451
1.7513	.1637	.7165	.6680	.2570	.9186	.8838
2.3833	.1709	.6215	.5779	.4084	.8706	.8120
3.0177	.1736	.5452	.5366	.5249	.8337	.7507
3.6893	.1749	.4800	.5179	.6183	.8042	.6952
4.3732	.1763	.4263	.5135	.6877	.7822	.6465
5.1393	.1786	.3778	.5187	.7415	.7651	.5995
5.8095	.1815	.3432	.5281	.7726	.7553	.5636
6.4663	.1852	.3147	.5398	.7922	.7491	.5324
7.1084	.1895	.2912	.5522	.8038	.7454	.5051
7.8141	.1949	.2693	.5661	.8105	.7433	.4781
8.7354	.2029	.2458	.5836	.8128	.7425	.4469
9.6517	.2114	.2267	.5995	.8112	.7430	.4197
10.5880	.2204	.2108	.6135	.8085	.7439	.3951
11.5738	.2297	.1970	.6255	.8065	.7445	.3721
12.5498	.2383	.1857	.6349	.8062	.7446	.3519
13.7320	.2478	.1743	.6434	.8084	.7439	.3301
15.0835	.2574	.1638	.6502	.8135	.7423	.3083
16.6517	.2667	.1540	.6554	.8218	.7397	.2864
18.4995	.2758	.1448	.6592	.8331	.7361	.2643
20.6802	.2844	.1365	.6618	.8468	.7318	.2422
23.2204	.2921	.1290	.6634	.8622	.7269	.2207
26.0155	.2985	.1228	.6643	.8778	.7219	.2010
29.0761	.3037	.1176	.6647	.8929	.7172	.1832
32.4317	.3079	.1133	.6649	.9071	.7127	.1669
36.1109	.3113	.1096	.6650	.9201	.7085	.1521
40.1434	.3141	.1066	.6652	.9317	.7049	.1387
44.5614	.3163	.1040	.6655	.9420	.7016	.1264
49.4025	.3181	.1019	.6659	.9509	.6988	.1152
54.7110	.3195	.1001	.6664	.9585	.6964	.1050
60.5373	.3207	.0985	.6671	.9651	.6943	.0958
66.3814	.3216	.0973	.6677	.9702	.6927	.0880
73.3630	.3225	.0962	.6685	.9750	.6912	.0802
81.0417	.3232	.0953	.6693	.9790	.6899	.0730
89.4888	.3237	.0945	.6701	.9824	.6888	.0665
98.7812	.3242	.0938	.6709	.9852	.6879	.0606
109.0019	.3246	.0933	.6717	.9875	.6872	.0552
120.2407	.3250	.0928	.6726	.9893	.6866	.0503
132.5952	.3254	.0925	.6734	.9908	.6861	.0458
146.1712	.3257	.0922	.6742	.9920	.6858	.0417
161.0841	.3260	.0919	.6750	.9928	.6855	.0379
177.4596	.3264	.0918	.6757	.9935	.6853	.0345
200.1960	.3268	.0916	.6766	.9942	.6851	.0307

MACH NO = 15.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0447	1.0391
.9650	.1399	.8644	1.0412	-.0197	1.0063	.9931
1.3316	.1525	.7864	.7960	.1275	.9596	.9390
1.8596	.1629	.6920	.6441	.2881	.9087	.8706
2.4189	.1675	.6101	.5707	.4197	.8671	.8083
3.1157	.1690	.5280	.5271	.5467	.8268	.7421
3.8432	.1692	.4601	.5091	.6449	.7957	.6836
4.5702	.1697	.4059	.5065	.7147	.7736	.6337
5.2029	.1710	.3672	.5116	.7571	.7602	.5959
5.8922	.1734	.3322	.5219	.7880	.7504	.5595
6.5541	.1766	.3041	.5344	.8064	.7446	.5285
7.1188	.1800	.2835	.5461	.8154	.7417	.5046
7.7288	.1843	.2645	.5590	.8200	.7403	.4812
8.6347	.1919	.2407	.5782	.8196	.7404	.4501
9.6479	.1996	.2233	.5944	.8151	.7418	.4254
10.2079	.2077	.2088	.6088	.8095	.7436	.4034
11.1538	.2167	.1953	.6222	.8041	.7453	.3816
12.0366	.2252	.1842	.6327	.8009	.7463	.3623
13.0811	.2345	.1731	.6421	.8001	.7466	.3418
14.1724	.2432	.1635	.6491	.8024	.7458	.3227
15.4018	.2516	.1545	.6546	.8077	.7442	.3036
16.9473	.2607	.1452	.6590	.8165	.7413	.2826
18.6688	.2691	.1370	.6621	.8276	.7378	.2624
20.8728	.2777	.1287	.6643	.8421	.7333	.2404
23.3152	.2851	.1216	.6655	.8575	.7284	.2199
26.1316	.2915	.1153	.6660	.8741	.7231	.2003
29.4767	.2970	.1097	.6660	.8914	.7176	.1811
32.9021	.3011	.1053	.6659	.9062	.7129	.1649
36.9660	.3047	.1013	.6659	.9205	.7084	.1491
41.1164	.3074	.0983	.6660	.9321	.7047	.1358
45.6650	.3096	.0957	.6663	.9422	.7015	.1237
51.0589	.3116	.0934	.6667	.9516	.6986	.1118
56.5826	.3131	.0916	.6673	.9590	.6962	.1019
62.6611	.3143	.0901	.6679	.9653	.6942	.0928
69.8974	.3154	.0887	.6687	.9710	.6924	.0839
77.3269	.3163	.0876	.6695	.9754	.6910	.0763
86.1746	.3172	.0867	.6704	.9793	.6898	.0690
95.2575	.3179	.0859	.6712	.9822	.6889	.0627
105.2599	.3185	.0853	.6721	.9846	.6881	.0571
117.1631	.3192	.0848	.6731	.9867	.6875	.0515
129.3706	.3198	.0844	.6739	.9882	.6870	.0469
143.8850	.3204	.0840	.6748	.9894	.6866	.0423
158.7562	.3209	.0838	.6756	.9903	.6863	.0385
175.0992	.3215	.0836	.6764	.9911	.6861	.0350
200.4188	.3222	.0833	.6774	.9919	.6858	.0307

MACH NO = 20.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1295	.9279	1.3710	-.1405	1.0445	1.0388
.9640	.1396	.8619	1.0413	-.0198	1.0063	.9931
1.3293	.1519	.7845	.7971	.1266	.9599	.9394
1.8511	.1619	.6908	.6448	.2866	.9092	.8716
2.4777	.1664	.5999	.5640	.4333	.8627	.8022
3.1699	.1672	.5200	.5229	.5576	.8234	.7374
3.8859	.1673	.4541	.5060	.6531	.7931	.6805
4.5965	.1672	.4017	.5037	.7210	.7716	.6321
5.2868	.1683	.3601	.5096	.7664	.7572	.5912
5.9496	.1705	.3270	.5199	.7950	.7482	.5567
6.5820	.1734	.3004	.5322	.8118	.7428	.5273
7.1181	.1765	.2810	.5434	.8200	.7402	.5047
7.6939	.1804	.2629	.5560	.8240	.7390	.4825
8.6004	.1879	.2389	.5750	.8227	.7394	.4512
9.4085	.1955	.2214	.5929	.8169	.7412	.4266
10.2055	.2036	.2070	.6079	.8100	.7434	.4048
11.0157	.2120	.1946	.6208	.8037	.7454	.3848
11.8703	.2204	.1836	.6318	.7994	.7458	.3657
12.8002	.2290	.1734	.6409	.7975	.7474	.3470
13.8368	.2376	.1638	.6484	.7987	.7470	.3283
14.9957	.2460	.1548	.6541	.8031	.7456	.3097
16.3341	.2543	.1462	.6595	.8105	.7433	.2906
18.0471	.2633	.1373	.6621	.8214	.7398	.2694
19.8940	.2712	.1296	.6644	.8336	.7359	.2497
22.0603	.2787	.1225	.6658	.8477	.7315	.2300
24.7237	.2856	.1157	.6664	.8645	.7262	.2096
27.8192	.2915	.1097	.6664	.8822	.7206	.1901
31.2370	.2962	.1046	.6662	.8986	.7153	.1724
35.0093	.3000	.1004	.6661	.9134	.7107	.1563
39.1610	.3032	.0969	.6661	.9264	.7065	.1417
44.0710	.3059	.0938	.6664	.9384	.7028	.1276
49.1360	.3079	.0913	.6668	.9479	.6997	.1158
54.7273	.3097	.0893	.6673	.9561	.6971	.1050
60.9099	.3111	.0876	.6679	.9630	.6950	.0952
67.7516	.3123	.0862	.6687	.9688	.6931	.0863
75.3245	.3134	.0850	.6695	.9736	.6916	.0782
83.7060	.3143	.0840	.6704	.9775	.6904	.0709
92.9813	.3151	.0832	.6713	.9807	.6893	.0642
104.0164	.3160	.0824	.6723	.9834	.6885	.0577
115.4475	.3167	.0819	.6733	.9854	.6879	.0522
128.0837	.3174	.0814	.6742	.9870	.6874	.0473
142.0448	.3181	.0811	.6751	.9882	.6870	.0428
157.4615	.3188	.0808	.6759	.9892	.6867	.0388
174.4774	.3194	.0806	.6767	.9900	.6864	.0351
200.4135	.3203	.0804	.6777	.9908	.6861	.0307

MACH NO = 25.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	YVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0444	1.0386
.9648	.1395	.9609	1.0413	-.0198	1.0063	.9932
1.3268	.1517	.7838	.7976	.1262	.9600	.9396
1.8472	.1614	.6905	.6451	.2859	.9094	.8721
2.4703	.1658	.5999	.5639	.4326	.8630	.8030
3.1579	.1664	.5202	.5223	.5569	.8236	.7384
3.8680	.1660	.4546	.5050	.6527	.7932	.6818
4.5717	.1650	.4022	.5022	.7211	.7716	.6336
5.2539	.1639	.3608	.5077	.7669	.7571	.5930
5.9074	.1628	.3278	.5177	.7959	.7479	.5587
6.5292	.1715	.3014	.5297	.8132	.7424	.5296
7.1190	.1748	.2799	.5421	.8224	.7395	.5046
7.6787	.1786	.2622	.5545	.8261	.7383	.4830
8.5565	.1857	.2388	.5742	.8245	.7388	.4526
9.3893	.1936	.2207	.5921	.8179	.7409	.4271
10.1517	.2014	.2068	.6068	.8106	.7432	.4062
10.9792	.2100	.1940	.6204	.8035	.7455	.3857
11.7893	.2181	.1834	.6312	.7988	.7470	.3675
12.7333	.2270	.1729	.6409	.7964	.7477	.3483
13.7123	.2352	.1637	.6482	.7972	.7475	.3305
14.8815	.2438	.1544	.6541	.8015	.7461	.3114
16.1396	.2517	.1462	.6584	.8083	.7439	.2932
17.7345	.2603	.1376	.6619	.8184	.7408	.2730
19.4216	.2679	.1303	.6643	.8295	.7372	.2544
21.5187	.2756	.1230	.6659	.8433	.7329	.2346
23.9038	.2823	.1165	.6666	.8587	.7280	.2155
27.0525	.2888	.1098	.6666	.8775	.7220	.1946
30.3405	.2936	.1046	.6664	.8943	.7167	.1767
34.2387	.2978	.0999	.6662	.9105	.7116	.1593
38.2508	.3010	.0963	.6662	.9236	.7074	.1447
42.9874	.3038	.0931	.6664	.9358	.7036	.1305
47.8650	.3060	.0906	.6668	.9455	.7005	.1186
53.6465	.3079	.0884	.6673	.9544	.6977	.1069
59.6262	.3094	.0866	.6679	.9614	.6955	.0971
66.7331	.3107	.0850	.6687	.9677	.6935	.0875
74.0911	.3118	.0838	.6695	.9725	.6919	.0794
82.8363	.3129	.0827	.6705	.9767	.6906	.0716
91.8893	.3138	.0819	.6714	.9799	.6896	.0649
102.6430	.3146	.0811	.6724	.9827	.6887	.0584
113.7682	.3154	.0806	.6733	.9847	.6881	.0530
126.9742	.3162	.0801	.6743	.9863	.6876	.0477
140.6204	.3170	.0797	.6752	.9876	.6872	.0432
156.8003	.3177	.0794	.6761	.9886	.6868	.0389
173.5013	.3184	.0792	.6769	.9894	.6866	.0353
200.3467	.3193	.0790	.6779	.9903	.6863	.0307

MACH NO = 30.00 CONE ANGLE = 9.00 ANGLE OF ATTACK = 10.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0444	1.0385
1.0046	.1411	.8512	1.0034	-.0017	1.0005	.9870
1.3779	.1529	.7731	.7758	.1440	.9544	.9326
1.9096	.1619	.6799	.6335	.3026	.9042	.8647
2.5400	.1656	.5903	.5576	.4472	.8583	.7959
3.2294	.1659	.5123	.5191	.5686	.8199	.7323
3.9366	.1654	.4483	.5035	.6614	.7905	.6768
4.6345	.1654	.3975	.5016	.7272	.7696	.6297
5.3092	.1663	.3571	.5076	.7712	.7557	.5900
5.9541	.1682	.3251	.5177	.7989	.7469	.5564
6.5666	.1708	.2993	.5296	.8153	.7417	.5279
7.1465	.1741	.2783	.5420	.8240	.7390	.5035
7.7554	.1782	.2593	.5556	.8274	.7379	.4802
8.5564	.1847	.2381	.5738	.8253	.7386	.4526
9.3710	.1925	.2204	.5915	.8185	.7407	.4277
10.1677	.2006	.2059	.6071	.8105	.7433	.4058
10.9736	.2091	.1935	.6205	.8033	.7455	.3858
11.7612	.2170	.1831	.6311	.7984	.7471	.3681
12.6768	.2257	.1729	.6407	.7958	.7479	.3494
13.6937	.2343	.1632	.6483	.7965	.7477	.3308
14.8277	.2426	.1542	.6542	.8006	.7464	.3123
16.0442	.2504	.1461	.6584	.8072	.7443	.2945
17.5776	.2587	.1378	.6619	.8169	.7412	.2749
19.3045	.2666	.1301	.6643	.8282	.7376	.2556
21.3026	.2741	.1231	.6660	.8414	.7335	.2365
23.5580	.2807	.1167	.6668	.8561	.7288	.2181
26.5788	.2871	.1101	.6668	.8746	.7230	.1975
30.0316	.2924	.1044	.6664	.8927	.7172	.1782
33.8635	.2966	.0996	.6662	.9090	.7121	.1608
37.8043	.2998	.0960	.6662	.9222	.7079	.1462
42.4520	.3027	.0927	.6664	.9345	.7040	.1320
47.5931	.3050	.0900	.6668	.9449	.7007	.1192
53.2958	.3070	.0877	.6673	.9537	.6979	.1076
59.1903	.3085	.0860	.6679	.9607	.6957	.0978
66.1910	.3099	.0844	.6687	.9670	.6937	.0882
73.9788	.3111	.0831	.6696	.9722	.6920	.0795
82.6411	.3121	.0820	.6705	.9764	.6907	.0717
92.2748	.3131	.0811	.6715	.9798	.6896	.0646
102.2364	.3139	.0804	.6724	.9823	.6888	.0587
114.0577	.3148	.0798	.6734	.9844	.6882	.0529
127.1890	.3156	.0793	.6744	.9861	.6876	.0476
141.7663	.3164	.0789	.6753	.9873	.6872	.0429
156.8078	.3171	.0786	.6762	.9883	.6869	.0389
174.6178	.3179	.0784	.6770	.9891	.6867	.0351
201.4006	.3189	.0782	.6780	.9900	.6864	.0305

MACH NO = 3.50 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1669	1.0588	1.0542
.8518	.1395	.9803	1.1745	-.0751	1.0265	1.0108
1.0594	.1533	.9269	.9637	.0188	.9934	.9747
1.3793	.1718	.8547	.7956	.1303	.9541	.9239
1.7730	.1896	.7796	.6951	.2347	.9172	.8683
2.2338	.2048	.7067	.6362	.3295	.8838	.8110
2.8538	.2187	.6276	.5975	.4279	.8491	.7450
3.4504	.2284	.5673	.5811	.4993	.8239	.6908
4.1024	.2370	.5146	.5748	.5582	.8032	.6400
4.8070	.2451	.4691	.5752	.6053	.7865	.5929
5.5638	.2530	.4302	.5795	.6427	.7733	.5494
6.3758	.2605	.3971	.5858	.6726	.7628	.5093
7.4017	.2690	.3646	.5941	.7006	.7529	.4664
8.5268	.2770	.3374	.6027	.7231	.7450	.4269
9.9397	.2857	.3120	.6120	.7440	.7376	.3858
11.4612	.2935	.2919	.6203	.7608	.7317	.3496
12.8882	.2996	.2778	.6266	.7734	.7273	.3214
14.6420	.3058	.2647	.6327	.7861	.7228	.2923
16.5440	.3111	.2542	.6379	.7974	.7188	.2662
18.3443	.3152	.2466	.6419	.8054	.7156	.2455
20.5743	.3192	.2395	.6458	.8158	.7123	.2239
23.0076	.3226	.2336	.6491	.8244	.7093	.2043
25.3200	.3251	.2294	.6517	.8313	.7068	.1885
28.1923	.3275	.2254	.6544	.8385	.7043	.1721
31.3325	.3295	.2221	.6568	.8450	.7020	.1571
34.3201	.3310	.2196	.6587	.8500	.7002	.1451
38.0335	.3324	.2173	.6606	.8553	.6984	.1325
42.0949	.3335	.2154	.6625	.8599	.6967	.1210
45.9601	.3343	.2140	.6639	.8635	.6955	.1118
50.7658	.3350	.2127	.6655	.8672	.6942	.1021
56.0234	.3356	.2116	.6670	.8704	.6930	.0933
61.0293	.3359	.2107	.6682	.8729	.6922	.0862
67.2531	.3362	.2099	.6695	.8753	.6913	.0788
74.0661	.3364	.2093	.6707	.8774	.6906	.0719
80.5542	.3366	.2088	.6717	.8790	.6900	.0665
88.6268	.3367	.2083	.6728	.8805	.6895	.0607
97.4655	.3367	.2078	.6738	.8818	.6890	.0555
105.8855	.3367	.2075	.6747	.8828	.6887	.0513
116.3643	.3367	.2072	.6756	.8837	.6884	.0468
127.8399	.3366	.2070	.6765	.8845	.6881	.0428
138.7732	.3366	.2068	.6772	.8850	.6879	.0395
152.3810	.3365	.2066	.6780	.8855	.6877	.0361
167.2833	.3364	.2065	.6787	.8859	.6876	.0330
181.4809	.3364	.2063	.6793	.8862	.6875	.0305
201.4737	.3363	.2062	.6800	.8864	.6874	.0275

MACH NO = 5.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0539	1.0457
.8768	.1387	.9271	1.1418	-.0626	1.0221	1.0063
1.1386	.1523	.8621	.9068	.0510	.9820	.9617
1.5157	.1674	.7813	.7438	.1755	.9381	.9039
1.9693	.1797	.7002	.6513	.2894	.8979	.8429
2.4939	.1889	.6234	.5991	.3909	.8621	.7820
3.0768	.1957	.5545	.5716	.4770	.8318	.7238
3.7036	.2015	.4949	.5604	.5456	.8076	.6702
4.3622	.2074	.4447	.5596	.5973	.7894	.6218
5.0452	.2135	.4028	.5648	.6351	.7760	.5785
5.7510	.2200	.3678	.5732	.6623	.7664	.5396
6.3755	.2259	.3423	.5815	.6795	.7604	.5094
7.1300	.2330	.3169	.5916	.6945	.7551	.4770
8.2658	.2436	.2868	.6056	.7098	.7497	.4354
9.3630	.2533	.2647	.6170	.7201	.7461	.4016
10.5563	.2629	.2460	.6270	.7291	.7429	.3703
11.8782	.2724	.2301	.6355	.7380	.7398	.3409
13.3677	.2815	.2164	.6425	.7475	.7364	.3129
15.0638	.2901	.2045	.6482	.7580	.7327	.2861
16.9237	.2978	.1947	.6526	.7690	.7288	.2616
19.1755	.3050	.1859	.6561	.7812	.7245	.2370
21.3898	.3105	.1795	.6585	.7920	.7207	.2169
23.8009	.3152	.1741	.6604	.8025	.7170	.1986
26.4280	.3190	.1697	.6619	.8125	.7135	.1818
29.2915	.3221	.1660	.6631	.8219	.7102	.1666
32.4136	.3247	.1629	.6641	.8306	.7071	.1526
36.2146	.3269	.1600	.6651	.8393	.7040	.1384
39.9625	.3284	.1579	.6660	.8463	.7015	.1268
44.0498	.3296	.1561	.6669	.8525	.6994	.1162
48.5077	.3306	.1546	.6678	.8580	.6974	.1065
53.3708	.3313	.1534	.6687	.8627	.6958	.0976
58.6770	.3319	.1523	.6696	.8667	.6944	.0894
64.4683	.3323	.1514	.6705	.8702	.6931	.0819
71.5279	.3327	.1505	.6715	.8734	.6920	.0743
78.4993	.3329	.1499	.6724	.8758	.6911	.0681
86.1133	.3331	.1493	.6733	.8779	.6904	.0624
94.4303	.3332	.1489	.6742	.8796	.6898	.0572
103.5159	.3332	.1485	.6750	.8811	.6893	.0524
113.4414	.3332	.1482	.6758	.8823	.6889	.0480
124.2842	.3332	.1479	.6766	.8832	.6885	.0439
137.5099	.3332	.1477	.6774	.8841	.6882	.0399
150.5737	.3332	.1475	.6782	.8846	.6880	.0365
164.8405	.3332	.1473	.6789	.8851	.6879	.0334
180.4189	.3332	.1472	.6795	.8854	.6878	.0306
201.4136	.3332	.1471	.6803	.8856	.6877	.0275

MACH NO = 10.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0504	1.0401
.9619	.1402	.8690	1.0462	-.0220	1.0078	.9914
1.2624	.1514	.7984	.8345	.0984	.9653	.9419
1.7440	.1625	.7030	.6782	.2447	.9137	.8721
2.2523	.1684	.6210	.6025	.3621	.8723	.8089
2.8152	.1713	.5472	.5606	.4632	.8367	.7488
3.4821	.1732	.4773	.5392	.5521	.8053	.6882
4.0821	.1750	.4268	.5346	.6093	.7851	.6415
4.6773	.1774	.3857	.5380	.6494	.7710	.6010
5.3348	.1811	.3483	.5471	.6788	.7606	.5619
5.9061	.1850	.3213	.5574	.6950	.7549	.5318
6.4646	.1894	.2989	.5685	.7048	.7515	.5053
7.0788	.1949	.2779	.5809	.7106	.7494	.4791
7.8815	.2028	.2552	.5967	.7131	.7485	.4487
8.7472	.2120	.2354	.6122	.7123	.7488	.4199
9.5647	.2207	.2202	.6246	.7107	.7494	.3959
10.4240	.2296	.2069	.6354	.7099	.7496	.3735
11.4332	.2394	.1942	.6450	.7108	.7493	.3503
12.4718	.2484	.1834	.6522	.7138	.7483	.3291
13.6523	.2573	.1735	.6580	.7191	.7464	.3080
15.1380	.2668	.1634	.6627	.7277	.7434	.2850
16.7533	.2752	.1549	.6657	.7382	.7397	.2636
18.6518	.2832	.1470	.6678	.7505	.7353	.2423
21.0927	.2910	.1392	.6691	.7656	.7300	.2194
23.6592	.2971	.1331	.6696	.7800	.7249	.1996
26.7205	.3025	.1277	.6698	.7947	.7197	.1802
29.8347	.3064	.1235	.6699	.8074	.7153	.1639
33.2538	.3096	.1200	.6699	.8188	.7113	.1492
37.3353	.3123	.1169	.6701	.8297	.7074	.1347
41.4860	.3144	.1145	.6704	.8385	.7043	.1226
46.0429	.3160	.1125	.6708	.8461	.7016	.1116
51.4886	.3174	.1107	.6713	.8531	.6991	.1008
57.0379	.3185	.1093	.6720	.8586	.6972	.0918
63.1444	.3193	.1081	.6727	.8632	.6956	.0835
70.4573	.3201	.1071	.6735	.8673	.6941	.0754
77.9197	.3207	.1063	.6743	.8705	.6930	.0686
86.8580	.3213	.1056	.6751	.8733	.6920	.0619
95.9776	.3217	.1050	.6760	.8754	.6913	.0563
106.0157	.3221	.1046	.6768	.8771	.6907	.0512
118.0304	.3225	.1042	.6777	.8786	.6902	.0462
130.2779	.3228	.1039	.6785	.8796	.6898	.0420
143.7462	.3232	.1037	.6792	.8804	.6895	.0382
159.8507	.3235	.1035	.6800	.8811	.6893	.0345
176.2518	.3238	.1034	.6807	.8816	.6891	.0313
200.6665	.3242	.1032	.6815	.8821	.6889	.0276

MACH NO = 15.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0498	1.0391
.9619	.1396	.8620	1.0460	-.0219	1.0077	.9914
1.3028	.1512	.7828	.8144	.1131	.9601	.9356
1.7832	.1607	.6893	.6678	.2568	.9095	.8669
2.3504	.1656	.6002	.5889	.3854	.8641	.7977
2.9726	.1672	.5223	.5482	.4926	.8263	.7335
3.5428	.1678	.4647	.5321	.5662	.8003	.6832
4.1811	.1689	.4123	.5280	.6260	.7792	.6344
4.8040	.1708	.3706	.5326	.6659	.7652	.5931
5.4056	.1737	.3374	.5418	.6910	.7563	.5580
5.9839	.1773	.3106	.5531	.7059	.7510	.5279
6.4773	.1810	.2909	.5636	.7134	.7484	.5047
7.0140	.1855	.2724	.5754	.7174	.7470	.4817
7.8131	.1933	.2494	.5929	.7175	.7470	.4511
8.5896	.2016	.2311	.6086	.7141	.7482	.4249
9.3675	.2103	.2160	.6225	.7099	.7496	.4015
10.1125	.2186	.2039	.6334	.7068	.7507	.3814
10.9658	.2275	.1923	.6434	.7052	.7513	.3607
11.9046	.2365	.1816	.6515	.7059	.7510	.3403
12.9601	.2454	.1717	.6580	.7094	.7498	.3201
14.0676	.2535	.1630	.6625	.7151	.7478	.3012
15.4557	.2621	.1542	.6661	.7239	.7447	.2806
17.1154	.2706	.1458	.6687	.7352	.7407	.2593
19.0666	.2785	.1381	.6702	.7483	.7361	.2380
21.2422	.2853	.1313	.6710	.7622	.7312	.2181
24.0296	.2918	.1248	.6711	.7784	.7255	.1970
27.1542	.2971	.1193	.6710	.7938	.7201	.1777
30.6179	.3013	.1147	.6709	.8077	.7152	.1603
34.1655	.3045	.1112	.6709	.8192	.7111	.1457
38.3797	.3072	.1081	.6710	.8300	.7073	.1315
43.0433	.3094	.1056	.6713	.8393	.7040	.1186
48.2122	.3112	.1035	.6717	.8472	.7012	.1071
53.5194	.3126	.1018	.6723	.8534	.6990	.0973
59.8459	.3138	.1004	.6730	.8590	.6971	.0878
66.8780	.3148	.0991	.6738	.8636	.6954	.0792
74.6948	.3157	.0981	.6746	.8674	.6941	.0714
82.7312	.3165	.0974	.6755	.8702	.6931	.0648
92.3126	.3172	.0967	.6764	.8727	.6922	.0584
102.9559	.3179	.0962	.6773	.8746	.6916	.0527
114.7743	.3185	.0957	.6782	.8762	.6910	.0474
127.8917	.3191	.0954	.6791	.8774	.6906	.0428
141.3547	.3196	.0951	.6799	.8783	.6903	.0388
157.3758	.3202	.0949	.6806	.8790	.6900	.0350
175.1380	.3207	.0947	.6814	.8797	.6898	.0315
200.8315	.3213	.0945	.6822	.8804	.6895	.0276

MACH NO = 20.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.0

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1285	.9279	1.3710	-.1405	1.0496	1.0388
.9629	.1395	.8594	1.0449	-.0214	1.0076	.9912
1.3075	.1510	.7734	.8119	.1149	.9595	.9349
1.7928	.1601	.6853	.6650	.2599	.9083	.8656
2.3637	.1645	.5960	.5803	.3894	.8627	.7963
2.9852	.1657	.5185	.5458	.4965	.8249	.7323
3.6216	.1660	.4552	.5287	.5776	.7963	.6767
4.2493	.1668	.4049	.5256	.6346	.7762	.6296
4.8574	.1686	.3650	.5306	.6724	.7629	.5898
5.4405	.1712	.3331	.5398	.6960	.7545	.5560
5.9974	.1745	.3074	.5509	.7100	.7496	.5273
6.4706	.1779	.2886	.5612	.7170	.7472	.5050
6.9854	.1821	.2709	.5727	.7204	.7459	.4831
7.7873	.1898	.2474	.5910	.7198	.7462	.4520
8.5113	.1976	.2302	.6065	.7157	.7476	.4274
9.2807	.2064	.2150	.6210	.7104	.7495	.4040
10.0153	.2146	.2028	.6326	.7062	.7509	.3839
10.8478	.2236	.1912	.6430	.7037	.7518	.3634
11.6971	.2320	.1812	.6509	.7036	.7519	.3446
12.7098	.2429	.1713	.6576	.7063	.7509	.3247
13.7723	.2490	.1626	.6624	.7115	.7491	.3060
14.9988	.2570	.1543	.6660	.7191	.7464	.2870
16.5447	.2654	.1459	.6697	.7296	.7427	.2662
18.1927	.2727	.1387	.6704	.7409	.7397	.2471
20.2883	.2802	.1315	.6714	.7547	.7338	.2264
22.7137	.2866	.1250	.6717	.7698	.7285	.2064
25.8221	.2926	.1187	.6715	.7867	.7226	.1855
29.0499	.2971	.1139	.6712	.8011	.7175	.1677
32.8851	.3010	.1096	.6711	.8147	.7127	.1507
36.8478	.3039	.1063	.6712	.8258	.7088	.1363
41.5498	.3064	.1034	.6714	.8359	.7052	.1225
46.4173	.3084	.1012	.6718	.8440	.7024	.1108
51.8061	.3099	.0993	.6724	.8509	.6999	.1003
58.2290	.3114	.0977	.6731	.8570	.6978	.0900
64.8996	.3125	.0964	.6738	.8617	.6961	.0814
72.8534	.3136	.0953	.6747	.8658	.6947	.0731
81.1124	.3144	.0945	.6756	.8689	.6936	.0660
90.9562	.3153	.0937	.6766	.8715	.6926	.0593
101.1720	.3161	.0932	.6775	.8735	.6920	.0535
113.3390	.3168	.0927	.6784	.8751	.6914	.0480
125.9545	.3175	.0923	.6793	.8763	.6910	.0434
140.9654	.3182	.0920	.6802	.8774	.6906	.0389
156.5160	.3188	.0918	.6809	.8781	.6903	.0352
173.7090	.3194	.0916	.6816	.8788	.6901	.0318
201.4613	.3201	.0914	.6825	.8796	.6898	.0275

MACH NO = 25.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC XCP/L	COEFFICIENTS YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0495	1.0386
.9629	.1394	.8584	1.0450	-.0215	1.0076	.9912
1.3061	.1508	.7787	.8124	.1146	.9596	.9351
1.7891	.1597	.6850	.6653	.2593	.9085	.8661
2.3570	.1639	.5959	.5862	.3887	.8629	.7970
2.9744	.1649	.5187	.5453	.4959	.8251	.7334
3.6057	.1651	.4556	.5277	.5773	.7964	.6780
4.2275	.1657	.4054	.5243	.6346	.7762	.6311
4.8287	.1672	.3655	.5289	.6728	.7627	.5915
5.4041	.1697	.3338	.5379	.6967	.7543	.5580
5.9522	.1728	.3082	.5488	.7110	.7493	.5295
6.4735	.1764	.2873	.5602	.7188	.7465	.5049
6.9701	.1804	.2700	.5716	.7220	.7454	.4835
7.7534	.1879	.2471	.5897	.7210	.7457	.4533
8.5024	.1960	.2292	.6061	.7162	.7474	.4277
9.2418	.2045	.2145	.6204	.7105	.7494	.4051
9.9958	.2130	.2019	.6326	.7058	.7511	.3844
10.7916	.2217	.1907	.6428	.7030	.7521	.3647
11.6000	.2298	.1811	.6506	.7025	.7523	.3467
12.5576	.2384	.1715	.6573	.7047	.7515	.3275
13.6305	.2467	.1625	.6623	.7097	.7497	.3084
14.8733	.2550	.1539	.6661	.7174	.7470	.2889
16.3235	.2630	.1458	.6688	.7272	.7435	.2690
17.8452	.2701	.1389	.6705	.7377	.7398	.2509
19.7529	.2773	.1319	.6716	.7505	.7353	.2314
22.1030	.2840	.1252	.6719	.7655	.7300	.2111
25.0479	.2901	.1188	.6717	.7823	.7241	.1903
28.3841	.2950	.1135	.6714	.7980	.7186	.1711
32.0933	.2990	.1091	.6712	.8119	.7137	.1539
35.9202	.3021	.1057	.6713	.8232	.7097	.1394
40.4538	.3047	.1027	.6715	.8335	.7061	.1254
45.4924	.3069	.1002	.6719	.8423	.7030	.1129
51.1040	.3086	.0982	.6724	.8497	.7003	.1015
57.3612	.3101	.0965	.6731	.8559	.6982	.0913
63.8516	.3113	.0952	.6739	.8607	.6965	.0827
71.5803	.3124	.0941	.6747	.8648	.6950	.0743
80.1995	.3134	.0931	.6757	.8682	.6938	.0668
89.8095	.3143	.0924	.6766	.8709	.6929	.0600
100.5205	.3151	.0917	.6776	.8730	.6922	.0539
112.4533	.3159	.0912	.6785	.8746	.6916	.0484
124.8095	.3167	.0909	.6794	.8758	.6912	.0438
139.4922	.3174	.0905	.6802	.8768	.6908	.0393
155.8264	.3181	.0903	.6810	.8776	.6905	.0353
173.9913	.3188	.0901	.6818	.8783	.6903	.0317
201.4063	.3196	.0899	.6826	.8791	.6900	.0275

MACH NO = 30.00 CONE ANGLE = 10.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0494	1.0385
1.0010	.1408	.9483	1.0088	-.0043	1.0015	.9846
1.3539	.1518	.7679	.7915	.1309	.9538	.9278
1.8463	.1601	.6744	.6541	.2742	.9033	.8586
2.4198	.1638	.5866	.5801	.4014	.8584	.7900
3.0381	.1645	.5110	.5421	.5059	.8216	.7274
3.6663	.1646	.4496	.5263	.5845	.7939	.6731
4.2830	.1652	.4007	.5237	.6397	.7744	.6272
4.8776	.1667	.3620	.5289	.6762	.7615	.5885
5.4457	.1691	.3310	.5380	.6991	.7534	.5558
5.9859	.1722	.3061	.5488	.7127	.7487	.5278
6.4991	.1758	.2857	.5602	.7200	.7461	.5038
7.0402	.1802	.2671	.5727	.7230	.7450	.4807
7.8061	.1876	.2450	.5906	.7213	.7456	.4514
8.5388	.1956	.2277	.6067	.7161	.7475	.4265
9.2620	.2039	.2134	.6208	.7103	.7495	.4045
9.9990	.2123	.2012	.6328	.7055	.7512	.3843
10.7760	.2208	.1903	.6428	.7026	.7522	.3651
11.6232	.2293	.1802	.6511	.7020	.7524	.3462
12.5591	.2377	.1708	.6576	.7042	.7517	.3275
13.6058	.2459	.1620	.6625	.7091	.7499	.3088
14.8155	.2539	.1537	.6661	.7166	.7473	.2897
16.2150	.2618	.1457	.6688	.7261	.7439	.2704
17.7830	.2692	.1385	.6706	.7368	.7401	.2516
19.6167	.2762	.1317	.6717	.7492	.7358	.2327
21.8625	.2827	.1252	.6721	.7637	.7307	.2130
24.7074	.2888	.1188	.6718	.7803	.7248	.1925
27.9751	.2938	.1133	.6715	.7962	.7192	.1733
31.6108	.2979	.1089	.6713	.8102	.7143	.1559
35.6408	.3012	.1052	.6713	.8223	.7100	.1404
40.1079	.3039	.1021	.6715	.8327	.7064	.1264
45.0698	.3061	.0996	.6719	.8415	.7033	.1138
50.5898	.3079	.0976	.6725	.8489	.7006	.1025
56.7418	.3094	.0959	.6731	.8551	.6984	.0922
63.5993	.3107	.0945	.6739	.8603	.6966	.0830
71.2423	.3118	.0933	.6748	.8645	.6951	.0746
79.7594	.3128	.0924	.6757	.8678	.6940	.0671
89.2484	.3137	.0916	.6766	.8705	.6930	.0603
99.8168	.3146	.0910	.6776	.8726	.6923	.0542
111.5820	.3154	.0905	.6786	.8742	.6917	.0487
124.6725	.3162	.0901	.6795	.8755	.6912	.0438
139.2305	.3170	.0897	.6803	.8765	.6909	.0394
155.4132	.3177	.0895	.6811	.8773	.6906	.0354
173.3955	.3184	.0893	.6819	.8780	.6904	.0318
200.5105	.3193	.0891	.6827	.8788	.6901	.0276

MACH NO = 3.50 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6834	.1273	1.0425	1.4632	-.1659	1.0894	1.0542
.8461	.1395	.9755	1.1892	-.0805	1.0432	1.0060
1.0519	.1535	.9047	.9947	.0027	.9986	.9531
1.3250	.1691	.8263	.8576	.0841	.9549	.8910
1.6343	.1829	.7537	.7749	.1526	.9182	.8297
1.9777	.1947	.6882	.7242	.2103	.8873	.7709
2.3511	.2045	.6306	.6928	.2584	.8615	.7157
2.7484	.2130	.5812	.6743	.2976	.8405	.6650
3.1656	.2208	.5394	.6646	.3286	.8239	.6190
3.6012	.2283	.5042	.6606	.3528	.8109	.5773
4.0565	.2357	.4744	.6601	.3718	.8008	.5393
4.4648	.2419	.4525	.6613	.3850	.7937	.5093
4.9654	.2488	.4304	.6636	.3941	.7867	.4767
5.7350	.2582	.4041	.6676	.4137	.7783	.4340
6.4952	.2662	.3846	.6714	.4256	.7719	.3988
7.3365	.2737	.3683	.6751	.4361	.7663	.3659
8.2796	.2807	.3545	.6784	.4458	.7611	.3349
9.3495	.2870	.3429	.6812	.4554	.7559	.3056
10.5747	.2926	.3332	.6834	.4648	.7509	.2777
12.1574	.2978	.3242	.6855	.4750	.7455	.2485
13.7844	.3017	.3177	.6870	.4835	.7409	.2242
15.5803	.3047	.3126	.6883	.4913	.7367	.2023
17.5646	.3070	.3085	.6895	.4982	.7330	.1827
19.7583	.3087	.3053	.6907	.5041	.7298	.1650
22.4696	.3102	.3025	.6919	.5098	.7268	.1473
25.1842	.3111	.3005	.6931	.5143	.7244	.1331
28.1882	.3117	.2989	.6942	.5180	.7224	.1202
31.5128	.3121	.2976	.6954	.5211	.7207	.1086
35.1929	.3124	.2965	.6965	.5237	.7194	.0981
39.2669	.3125	.2957	.6977	.5258	.7182	.0886
44.3077	.3126	.2950	.6990	.5277	.7172	.0791
49.3591	.3126	.2945	.7001	.5289	.7165	.0715
54.9527	.3126	.2940	.7012	.5300	.7160	.0646
61.1469	.3125	.2937	.7023	.5308	.7156	.0583
68.0060	.3125	.2934	.7033	.5313	.7153	.0527
76.4940	.3124	.2932	.7043	.5318	.7150	.0470
84.9997	.3124	.2930	.7052	.5321	.7149	.0425
94.4170	.3124	.2929	.7061	.5323	.7147	.0384
104.8429	.3124	.2928	.7069	.5324	.7147	.0347
116.3849	.3124	.2927	.7076	.5325	.7146	.0313
129.1614	.3124	.2927	.7083	.5325	.7146	.0283
144.9660	.3124	.2926	.7090	.5325	.7146	.0252
160.7978	.3124	.2926	.7095	.5325	.7146	.0228
178.3211	.3124	.2925	.7100	.5325	.7146	.0206
202.3005	.3125	.2925	.7106	.5325	.7146	.0192

MACH NO = 5.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XYCP/LV	RN/RB
.7077	.1284	.9838	1.4130	-.1528	1.0819	1.0457
.8944	.1399	.9111	1.1312	-.0583	1.0312	.9931
1.1217	.1517	.8360	.9460	.0283	.9848	.9364
1.4204	.1635	.7539	.8176	.1128	.9395	.8711
1.7560	.1731	.6787	.7414	.1834	.9017	.8079
2.1198	.1810	.6127	.6968	.2407	.8710	.7489
2.5024	.1877	.5565	.6714	.2859	.8468	.6955
2.8952	.1940	.5097	.6588	.3201	.8285	.6481
3.2941	.2003	.4710	.6544	.3450	.8151	.6061
3.6986	.2070	.4387	.6548	.3631	.8054	.5687
4.1096	.2139	.4118	.6579	.3763	.7984	.5352
4.5291	.2208	.3890	.6621	.3863	.7930	.5048
4.9596	.2278	.3695	.6668	.3942	.7887	.4770
5.5763	.2373	.3471	.6732	.4029	.7841	.4422
6.2330	.2467	.3283	.6792	.4102	.7802	.4103
6.9461	.2559	.3125	.6844	.4170	.7765	.3804
7.7360	.2647	.2989	.6884	.4243	.7726	.3521
8.6267	.2730	.2870	.6914	.4324	.7693	.3248
9.6475	.2806	.2767	.6935	.4411	.7636	.2983
10.8330	.2876	.2676	.6948	.4504	.7586	.2725
12.2234	.2937	.2598	.6954	.4605	.7532	.2474
13.8637	.2989	.2530	.6956	.4707	.7477	.2231
15.7816	.3031	.2474	.6956	.4807	.7424	.2002
17.7219	.3060	.2433	.6956	.4890	.7379	.1813
20.0865	.3083	.2397	.6956	.4972	.7336	.1626
22.7197	.3100	.2369	.6958	.5042	.7298	.1459
25.6526	.3111	.2347	.6961	.5101	.7266	.1309
28.9202	.3119	.2330	.6966	.5151	.7240	.1174
32.5616	.3124	.2316	.6974	.5191	.7218	.1053
36.6207	.3127	.2305	.6982	.5223	.7201	.0945
41.1464	.3128	.2297	.6991	.5249	.7187	.0848
46.1930	.3130	.2290	.7002	.5268	.7177	.0761
51.8212	.3130	.2284	.7012	.5283	.7169	.0682
58.0978	.3131	.2280	.7023	.5295	.7162	.0612
64.4291	.3131	.2277	.7032	.5303	.7158	.0555
72.1572	.3131	.2275	.7042	.5309	.7155	.0497
80.7737	.3131	.2272	.7052	.5313	.7153	.0446
90.3793	.3132	.2271	.7061	.5316	.7151	.0400
101.0865	.3132	.2269	.7069	.5318	.7150	.0359
113.0202	.3133	.2268	.7077	.5320	.7149	.0322
126.3201	.3133	.2267	.7084	.5321	.7149	.0289
141.1416	.3134	.2267	.7090	.5322	.7148	.0259
157.6585	.3134	.2266	.7096	.5322	.7148	.0232
176.0645	.3135	.2266	.7101	.5322	.7148	.0209
200.5496	.3135	.2265	.7107	.5323	.7147	.0183

MACH NO = 10.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

INVISCID AERODYNAMIC COEFFICIENTS

L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7250	.1286	.9394	1.3793	-.1430	1.0766	1.0401
.9573	.1402	.8527	1.0662	-.0309	1.0166	.9767
1.2421	.1504	.7643	.8782	.0688	.9631	.9090
1.5686	.1580	.6812	.7681	.1531	.9179	.8420
1.9647	.1640	.6003	.6990	.2285	.8775	.7729
2.3339	.1681	.5397	.6655	.2803	.8498	.7180
2.7429	.1723	.4853	.6474	.3219	.8275	.6657
3.1037	.1765	.4461	.6420	.3474	.8138	.6254
3.4940	.1819	.4110	.6431	.3661	.8038	.5870
3.8361	.1871	.3852	.6473	.3769	.7980	.5570
4.2076	.1933	.3616	.6534	.3848	.7938	.5278
4.5362	.1990	.3437	.6595	.3894	.7913	.5043
4.8971	.2055	.3269	.6664	.3928	.7895	.4809
5.4025	.2148	.3073	.6756	.3956	.7880	.4515
5.9200	.2241	.2910	.6837	.3978	.7868	.4249
6.4627	.2333	.2771	.6903	.4004	.7854	.4002
7.0450	.2423	.2648	.6955	.4039	.7835	.3766
7.6837	.2510	.2538	.6995	.4085	.7811	.3538
8.3981	.2595	.2438	.7023	.4143	.7780	.3314
9.2121	.2676	.2346	.7040	.4214	.7742	.3091
10.2273	.2757	.2255	.7046	.4306	.7692	.2851
11.3468	.2826	.2176	.7045	.4403	.7640	.2626
12.6786	.2890	.2104	.7038	.4508	.7584	.2401
14.2771	.2947	.2039	.7029	.4618	.7525	.2177
16.2049	.2996	.1981	.7019	.4727	.7467	.1957
18.5210	.3037	.1931	.7010	.4832	.7410	.1745
21.1617	.3068	.1892	.7003	.4925	.7360	.1553
24.1251	.3090	.1862	.6999	.5005	.7318	.1383
27.4521	.3106	.1839	.6998	.5072	.7282	.1231
31.1896	.3117	.1821	.7000	.5127	.7252	.1096
35.3908	.3126	.1807	.7005	.5171	.7229	.0976
40.1148	.3132	.1796	.7011	.5206	.7210	.0868
45.4272	.3137	.1787	.7019	.5233	.7196	.0773
51.4011	.3141	.1780	.7027	.5254	.7184	.0688
58.1177	.3144	.1775	.7037	.5270	.7176	.0612
65.6679	.3147	.1771	.7046	.5282	.7169	.0545
74.1535	.3149	.1767	.7055	.5291	.7165	.0485
84.4131	.3152	.1765	.7065	.5298	.7161	.0428
95.2150	.3154	.1762	.7073	.5303	.7158	.0381
107.3495	.3156	.1761	.7080	.5307	.7156	.0339
120.9800	.3157	.1759	.7087	.5310	.7155	.0301
136.2906	.3159	.1758	.7093	.5312	.7153	.0268
153.4884	.3160	.1757	.7099	.5314	.7152	.0239
172.8065	.3161	.1756	.7104	.5316	.7151	.0212
201.1852	.3162	.1755	.7110	.5318	.7150	.0183

MACH NO = 15.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7283	.1286	.9310	1.3731	-.1412	1.0757	1.0391
.9849	.1406	.8364	1.0410	-.0196	1.0105	.9697
1.2680	.1497	.7500	.8653	.0772	.9587	.9033
1.6272	.1569	.6608	.7524	.1674	.9103	.8310
2.0152	.1617	.5839	.6900	.2390	.8719	.7649
2.4109	.1652	.5209	.6571	.2925	.8433	.7076
2.8000	.1688	.4707	.6422	.3301	.8231	.6589
3.1768	.1730	.4309	.6383	.3550	.8098	.6178
3.5395	.1778	.3991	.6405	.3708	.8013	.5828
3.8891	.1831	.3733	.6457	.3808	.7959	.5527
4.2275	.1887	.3520	.6521	.3871	.7926	.5263
4.5567	.1945	.3342	.6589	.3908	.7905	.5029
4.9114	.2010	.3178	.6665	.3931	.7893	.4800
5.3910	.2101	.2992	.6762	.3946	.7885	.4521
5.8780	.2193	.2837	.6846	.3957	.7879	.4269
6.3648	.2283	.2704	.6915	.3974	.7870	.4035
6.9248	.2371	.2587	.6969	.4001	.7856	.3813
7.5129	.2457	.2481	.7011	.4038	.7836	.3597
8.1658	.2541	.2384	.7040	.4089	.7809	.3384
8.9556	.2626	.2288	.7058	.4159	.7771	.3158
9.8094	.2700	.2204	.7065	.4240	.7728	.2945
10.7967	.2770	.2126	.7064	.4330	.7679	.2732
11.9417	.2835	.2054	.7058	.4428	.7627	.2521
13.3137	.2894	.1987	.7049	.4532	.7571	.2307
15.0006	.2949	.1924	.7038	.4641	.7513	.2089
17.1942	.3000	.1865	.7027	.4755	.7452	.1861
19.6816	.3040	.1818	.7018	.4857	.7397	.1655
22.5302	.3070	.1781	.7012	.4947	.7349	.1470
25.7357	.3092	.1753	.7009	.5023	.7308	.1305
29.3458	.3109	.1731	.7009	.5086	.7275	.1159
33.4148	.3121	.1714	.7011	.5137	.7247	.1029
38.0030	.3131	.1701	.7016	.5177	.7226	.0913
43.5450	.3138	.1690	.7024	.5211	.7207	.0804
49.4267	.3144	.1682	.7032	.5235	.7194	.0714
56.0580	.3150	.1676	.7041	.5254	.7185	.0633
63.5329	.3154	.1670	.7050	.5268	.7177	.0562
71.9568	.3159	.1666	.7059	.5278	.7172	.0499
81.4482	.3162	.1663	.7068	.5286	.7167	.0443
92.8997	.3166	.1660	.7076	.5293	.7164	.0390
105.0394	.3169	.1658	.7084	.5298	.7161	.0346
118.7125	.3172	.1656	.7090	.5302	.7159	.0307
134.1126	.3174	.1654	.7096	.5306	.7157	.0272
151.4579	.3176	.1653	.7101	.5309	.7155	.0242
170.9946	.3177	.1652	.7106	.5311	.7154	.0215
200.9377	.3179	.1651	.7112	.5314	.7152	.0183

MACH NO = 20.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7294	.1285	.9279	1.3710	-.1405	1.0753	1.0388
.9842	.1403	.8340	1.0415	-.0198	1.0106	.9699
1.2648	.1491	.7483	.8662	.0765	.9590	.9040
1.6204	.1560	.6597	.7530	.1665	.9108	.8323
2.0428	.1609	.5763	.6857	.2442	.8692	.7606
2.4328	.1641	.5150	.6545	.2961	.8413	.7046
2.8146	.1674	.4662	.6405	.3325	.8218	.6573
3.1828	.1714	.4275	.6370	.3566	.8089	.6172
3.5704	.1765	.3937	.6398	.3730	.8001	.5800
3.9085	.1816	.3690	.6451	.3822	.7952	.5511
4.2347	.1870	.3486	.6516	.3879	.7921	.5258
4.5510	.1926	.3315	.6584	.3912	.7903	.5033
4.8906	.1989	.3157	.6660	.3931	.7893	.4813
5.3481	.2077	.2977	.6757	.3941	.7888	.4545
5.8413	.2171	.2818	.6846	.3949	.7884	.4287
6.3533	.2264	.2682	.6919	.3963	.7876	.4049
6.8988	.2354	.2562	.6976	.3988	.7863	.3823
7.4545	.2438	.2461	.7017	.4022	.7845	.3617
8.1113	.2524	.2362	.7047	.4072	.7818	.3400
8.8517	.2605	.2270	.7065	.4139	.7782	.3186
9.7026	.2682	.2184	.7071	.4220	.7739	.2970
10.6094	.2749	.2109	.7071	.4305	.7693	.2770
11.7212	.2815	.2036	.7065	.4401	.7641	.2559
13.0500	.2876	.1967	.7056	.4505	.7586	.2345
14.5766	.2929	.1906	.7045	.4608	.7531	.2140
16.4963	.2979	.1849	.7035	.4715	.7473	.1928
18.7729	.3022	.1800	.7026	.4816	.7419	.1725
21.5775	.3057	.1758	.7018	.4912	.7368	.1527
24.6045	.3082	.1728	.7014	.4991	.7325	.1359
28.2469	.3102	.1702	.7012	.5062	.7287	.1200
32.3754	.3117	.1683	.7014	.5119	.7257	.1059
37.0576	.3129	.1668	.7019	.5164	.7233	.0935
42.0166	.3137	.1657	.7025	.5197	.7215	.0832
47.9927	.3145	.1648	.7033	.5225	.7200	.0734
54.7694	.3151	.1640	.7042	.5245	.7189	.0648
61.9433	.3157	.1635	.7051	.5260	.7181	.0576
70.5835	.3162	.1630	.7060	.5271	.7175	.0508
80.3744	.3167	.1626	.7069	.5280	.7170	.0448
91.4671	.3172	.1623	.7078	.5287	.7167	.0396
103.2009	.3175	.1620	.7085	.5293	.7164	.0352
117.3242	.3179	.1618	.7092	.5297	.7161	.0311
133.3221	.3181	.1616	.7098	.5302	.7159	.0274
151.4434	.3184	.1615	.7103	.5305	.7157	.0242
170.6115	.3186	.1613	.7108	.5308	.7156	.0215
201.5059	.3189	.1612	.7114	.5311	.7154	.0183

MACH NO = 25.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7299	.1286	.9266	1.3700	-.1402	1.0752	1.0386
.9752	.1399	.8360	1.0493	-.0234	1.0125	.9722
1.2729	.1491	.7450	.8624	.0790	.9577	.9022
1.6472	.1560	.6527	.7468	.1726	.9075	.8273
2.0480	.1604	.5742	.6846	.2454	.8685	.7598
2.4545	.1636	.5107	.6526	.2992	.8396	.7018
2.8174	.1667	.4646	.6396	.3335	.8213	.6569
3.2030	.1708	.4242	.6362	.3585	.8079	.6152
3.5726	.1757	.3922	.6391	.3738	.7997	.5798
3.9273	.1810	.3664	.6449	.3832	.7946	.5496
4.2692	.1866	.3453	.6518	.3888	.7916	.5232
4.5706	.1920	.3292	.6585	.3917	.7901	.5020
4.8949	.1980	.3141	.6659	.3933	.7892	.4810
5.3618	.2071	.2958	.6760	.3941	.7888	.4537
5.8348	.2161	.2806	.6847	.3947	.7885	.4290
6.3264	.2251	.2675	.6919	.3958	.7879	.4061
6.8501	.2339	.2558	.6976	.3980	.7867	.3842
7.4204	.2426	.2453	.7019	.4013	.7849	.3629
8.0535	.2510	.2356	.7050	.4061	.7824	.3418
8.8136	.2595	.2261	.7068	.4130	.7787	.3196
9.6357	.2670	.2176	.7074	.4208	.7745	.2986
10.5642	.2740	.2099	.7074	.4296	.7698	.2779
11.6303	.2805	.2027	.7069	.4389	.7648	.2575
12.8964	.2865	.1960	.7060	.4490	.7594	.2368
14.4454	.2921	.1896	.7049	.4596	.7537	.2156
16.2512	.2970	.1840	.7038	.4698	.7482	.1952
18.3628	.3012	.1793	.7029	.4795	.7430	.1758
20.9484	.3047	.1751	.7022	.4889	.7380	.1567
24.0748	.3076	.1716	.7017	.4975	.7334	.1385
27.6136	.3098	.1690	.7014	.5048	.7295	.1225
31.6202	.3114	.1670	.7016	.5107	.7263	.1082
36.1593	.3126	.1654	.7020	.5153	.7238	.0956
41.3023	.3136	.1642	.7026	.5190	.7219	.0845
47.1290	.3144	.1632	.7034	.5218	.7204	.0747
53.7292	.3152	.1624	.7043	.5239	.7192	.0659
61.2041	.3158	.1618	.7052	.5255	.7184	.0583
69.6677	.3164	.1613	.7061	.5267	.7177	.0515
79.2484	.3169	.1609	.7070	.5276	.7172	.0455
90.0913	.3174	.1606	.7078	.5284	.7168	.0401
102.3612	.3178	.1603	.7086	.5290	.7165	.0355
116.2448	.3182	.1600	.7093	.5295	.7163	.0313
131.9542	.3186	.1598	.7099	.5299	.7160	.0277
149.7298	.3188	.1597	.7104	.5303	.7158	.0245
169.8441	.3191	.1595	.7109	.5306	.7157	.0216
200.3156	.3194	.1593	.7115	.5308	.7156	.0184

MACH NO = 30.00 CONE ANGLE = 15.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.7302	.1286	.9258	1.3694	-.1401	1.0751	1.0385
.9751	.1398	.8354	1.0494	-.0234	1.0125	.9722
1.2723	.1490	.7445	.8626	.0789	.9577	.9023
1.6457	.1558	.6524	.7469	.1724	.9076	.8276
2.0455	.1601	.5740	.6845	.2453	.8686	.7602
2.4505	.1632	.5106	.6524	.2991	.8397	.7023
2.8119	.1662	.4646	.6393	.3334	.8213	.6576
3.1957	.1703	.4242	.6358	.3584	.8079	.6159
3.5630	.1751	.3922	.6387	.3738	.7997	.5807
3.9153	.1803	.3665	.6444	.3832	.7946	.5505
4.2544	.1859	.3454	.6513	.3889	.7916	.5243
4.5530	.1912	.3293	.6580	.3917	.7901	.5032
4.8738	.1972	.3143	.6654	.3933	.7893	.4823
5.3637	.2067	.2950	.6761	.3940	.7889	.4536
5.8303	.2157	.2800	.6848	.3945	.7886	.4293
6.3148	.2245	.2670	.6919	.3955	.7880	.4066
6.8303	.2333	.2555	.6976	.3976	.7869	.3850
7.4275	.2424	.2444	.7022	.4011	.7851	.3626
8.0527	.2507	.2349	.7052	.4058	.7825	.3419
8.7546	.2586	.2260	.7069	.4121	.7791	.3212
9.5558	.2661	.2176	.7076	.4198	.7750	.3005
10.4564	.2730	.2100	.7076	.4283	.7705	.2802
11.5541	.2798	.2024	.7071	.4380	.7653	.2589
12.7811	.2858	.1957	.7062	.4479	.7600	.2385
14.2752	.2913	.1894	.7051	.4584	.7544	.2178
16.0069	.2962	.1839	.7041	.4684	.7490	.1978
18.1484	.3006	.1789	.7031	.4785	.7436	.1776
20.6064	.3042	.1747	.7024	.4876	.7387	.1590
23.6425	.3072	.1712	.7018	.4963	.7340	.1408
27.1066	.3094	.1684	.7016	.5037	.7301	.1245
31.2874	.3112	.1662	.7016	.5101	.7266	.1093
35.7594	.3125	.1646	.7020	.5148	.7241	.0966
40.8235	.3135	.1634	.7026	.5186	.7221	.0854
46.5573	.3144	.1624	.7034	.5214	.7206	.0755
53.4816	.3152	.1615	.7043	.5237	.7193	.0662
60.8860	.3159	.1609	.7053	.5253	.7185	.0585
69.2649	.3165	.1604	.7062	.5265	.7178	.0517
78.7441	.3171	.1600	.7071	.5274	.7173	.0457
90.1807	.3176	.1596	.7079	.5282	.7169	.0401
102.3998	.3181	.1593	.7087	.5288	.7166	.0355
116.2179	.3185	.1590	.7094	.5293	.7163	.0313
131.8438	.3188	.1588	.7099	.5298	.7161	.0277
150.6929	.3191	.1587	.7105	.5302	.7159	.0243
170.8310	.3194	.1585	.7110	.5305	.7157	.0215
201.3180	.3198	.1583	.7116	.5306	.7157	.0183

MACH NO = 3.50 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC XCP/L	COEFFICIENTS YCP/D	XVCP/LV	RN/RD
.6580	.1252	1.0558	1.5198	-.1820	1.1325	1.0642
.8325	.1397	.9705	1.2272	-.0943	1.0686	.9968
1.0227	.1538	.9951	1.0562	-.0268	1.0195	.9325
1.2303	.1667	.8280	.9494	.0271	.9803	.8711
1.4811	.1792	.7623	.8737	.0755	.9451	.8069
1.7223	.1891	.7113	.8296	.1106	.9195	.7535
2.0029	.1988	.6632	.7977	.1418	.8968	.6997
2.2664	.2066	.6264	.7784	.1646	.8802	.6557
2.5705	.2148	.5919	.7649	.1847	.8655	.6113
2.8554	.2220	.5656	.7575	.1990	.8551	.5749
3.1865	.2298	.5409	.7526	.2119	.8457	.5376
3.5006	.2365	.5219	.7498	.2218	.8385	.5065
3.8712	.2436	.5037	.7478	.2315	.8315	.4741
4.3109	.2511	.4866	.7465	.2408	.8247	.4407
4.7925	.2583	.4720	.7459	.2491	.8187	.4091
5.2761	.2645	.4606	.7453	.2564	.8134	.3816
5.8711	.2706	.4497	.7444	.2645	.8075	.3525
6.5485	.2760	.4404	.7433	.2726	.8016	.3243
7.3288	.2807	.4324	.7421	.2806	.7958	.2969
8.1480	.2844	.4262	.7410	.2878	.7905	.2728
9.1963	.2878	.4204	.7398	.2956	.7848	.2471
10.4289	.2904	.4155	.7387	.3030	.7794	.2224
11.8791	.2922	.4116	.7379	.3099	.7744	.1991
13.4153	.2933	.4088	.7373	.3156	.7702	.1791
15.3772	.2940	.4063	.7371	.3210	.7664	.1588
17.6591	.2943	.4044	.7374	.3253	.7632	.1403
20.2724	.2944	.4030	.7380	.3287	.7607	.1238
22.9494	.2944	.4021	.7388	.3310	.7590	.1105
26.2687	.2943	.4013	.7399	.3329	.7577	.0975
30.0318	.2942	.4007	.7411	.3342	.7567	.0860
34.2982	.2941	.4003	.7424	.3351	.7561	.0759
38.6699	.2941	.4000	.7436	.3356	.7557	.0677
44.0915	.2940	.3997	.7448	.3360	.7554	.0597
50.2374	.2940	.3995	.7459	.3362	.7553	.0527
57.2041	.2940	.3994	.7470	.3363	.7552	.0465
64.3410	.2940	.3993	.7478	.3364	.7551	.0415
73.1893	.2940	.3992	.7487	.3364	.7551	.0366
83.2175	.2940	.3991	.7494	.3365	.7551	.0323
94.5829	.2940	.3990	.7501	.3365	.7551	.0285
106.2249	.2940	.3990	.7506	.3365	.7550	.0254
120.6580	.2940	.3990	.7511	.3365	.7550	.0224
137.0161	.2940	.3989	.7516	.3365	.7550	.0198
155.5561	.2940	.3989	.7520	.3365	.7550	.0174
176.5695	.2940	.3989	.7524	.3365	.7550	.0154
200.3866	.2940	.3989	.7527	.3365	.7550	.0136

MACH NO = 5.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1249	1.0086	1.5198	-.1820	1.1325	1.0642
.8554	.1397	.9129	1.1988	-.0841	1.0612	.9886
1.0571	.1504	.8338	1.0289	-.0141	1.0102	.9217
1.3008	.1616	.7563	.9154	.0469	.9659	.8520
1.5610	.1711	.6898	.8469	.0942	.9314	.7884
1.8292	.1794	.6347	.8059	.1300	.9054	.7321
2.0997	.1872	.5896	.7817	.1565	.8861	.6829
2.3731	.1947	.5523	.7676	.1763	.8717	.6394
2.6493	.2023	.5216	.7603	.1908	.8611	.6008
2.9299	.2100	.4959	.7568	.2017	.8532	.5661
3.2179	.2178	.4742	.7552	.2104	.8468	.5344
3.5158	.2251	.4557	.7548	.2177	.8415	.5051
3.8556	.2332	.4383	.7552	.2243	.8367	.4754
4.2163	.2413	.4232	.7558	.2303	.8323	.4475
4.6387	.2496	.4091	.7561	.2368	.8276	.4187
5.0661	.2568	.3977	.7558	.2431	.8230	.3931
5.5816	.2641	.3867	.7551	.2502	.8178	.3661
6.1670	.2710	.3770	.7539	.2577	.8124	.3396
6.7867	.2768	.3690	.7525	.2649	.8071	.3154
7.5674	.2823	.3613	.7504	.2734	.8010	.2895
8.4149	.2867	.3550	.7483	.2814	.7952	.2657
9.5046	.2906	.3492	.7458	.2904	.7886	.2404
10.8125	.2935	.3442	.7435	.2992	.7822	.2157
12.2510	.2954	.3405	.7416	.3068	.7767	.1938
14.1050	.2967	.3373	.7401	.3142	.7713	.1714
16.1053	.2974	.3350	.7393	.3200	.7671	.1522
18.6454	.2977	.3331	.7390	.3250	.7634	.1336
21.5168	.2977	.3318	.7394	.3287	.7607	.1172
24.5223	.2976	.3309	.7400	.3312	.7589	.1039
28.2158	.2976	.3302	.7410	.3330	.7576	.0912
32.0831	.2975	.3297	.7421	.3343	.7567	.0808
36.8362	.2974	.3293	.7434	.3351	.7561	.0709
42.2551	.2974	.3290	.7446	.3356	.7557	.0622
47.9290	.2974	.3287	.7457	.3359	.7555	.0551
54.8996	.2974	.3285	.7468	.3361	.7553	.0484
62.8432	.2974	.3284	.7478	.3362	.7553	.0424
71.1581	.2974	.3283	.7486	.3363	.7552	.0376
81.3733	.2975	.3282	.7494	.3363	.7552	.0330
92.0643	.2975	.3281	.7500	.3364	.7551	.0292
105.1987	.2975	.3280	.7506	.3364	.7551	.0256
120.1674	.2975	.3280	.7512	.3364	.7551	.0225
135.8341	.2976	.3279	.7516	.3365	.7551	.0199
155.0826	.2976	.3279	.7520	.3365	.7551	.0175
175.2294	.2976	.3279	.7524	.3365	.7551	.0155
202.0245	.2976	.3278	.7527	.3365	.7550	.0135

MACH NO = 10.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1245	.9718	1.5198	-.1820	1.1325	1.0642
.8857	.1380	.8629	1.1641	-.0711	1.0517	.9779
1.1463	.1492	.7647	.9734	.0136	.9901	.8949
1.4485	.1585	.6760	.8630	.0808	.9412	.8147
1.7346	.1656	.6100	.8079	.1251	.9089	.7510
2.0323	.1727	.5553	.7774	.1571	.8856	.6945
2.3155	.1801	.5137	.7635	.1775	.8708	.6481
2.5847	.1877	.4815	.7582	.1904	.8614	.6094
2.8266	.1949	.4574	.7571	.1985	.8555	.5784
3.0786	.2024	.4362	.7577	.2048	.8509	.5492
3.3249	.2098	.4188	.7593	.2095	.8475	.5235
3.5520	.2166	.4051	.7610	.2130	.8450	.5017
3.7955	.2236	.3923	.7628	.2163	.8426	.4804
4.5144	.2420	.3638	.7659	.2255	.8358	.4267
5.2924	.2576	.3431	.7658	.2359	.8282	.3807
6.1499	.2703	.3272	.7634	.2476	.8198	.3403
7.0983	.2802	.3150	.7598	.2596	.8110	.3045
8.1481	.2878	.3055	.7560	.2712	.8026	.2728
9.3129	.2935	.2981	.7524	.2819	.7948	.2445
10.6075	.2977	.2925	.7493	.2915	.7878	.2192
12.0488	.3006	.2882	.7467	.3001	.7815	.1966
13.6540	.3025	.2849	.7445	.3077	.7760	.1764
15.4424	.3037	.2825	.7428	.3141	.7713	.1582
17.4358	.3043	.2806	.7417	.3196	.7674	.1419
19.6039	.3045	.2792	.7410	.3240	.7642	.1276
22.0759	.3044	.2781	.7408	.3276	.7616	.1145
24.8320	.3042	.2773	.7409	.3304	.7595	.1027
27.9044	.3040	.2766	.7413	.3325	.7580	.0921
31.3297	.3037	.2761	.7420	.3340	.7569	.0826
35.1487	.3035	.2757	.7428	.3350	.7561	.0741
39.4066	.3033	.2754	.7436	.3358	.7556	.0665
44.1531	.3031	.2751	.7446	.3363	.7552	.0596
49.4437	.3029	.2749	.7455	.3366	.7550	.0535
55.3401	.3028	.2747	.7464	.3367	.7549	.0480
61.9114	.3027	.2745	.7472	.3368	.7548	.0430
69.2352	.3026	.2744	.7480	.3369	.7548	.0386
77.3978	.3026	.2743	.7487	.3369	.7548	.0346
86.4955	.3025	.2742	.7493	.3369	.7548	.0311
96.6355	.3025	.2741	.7499	.3369	.7548	.0279
107.9371	.3024	.2741	.7504	.3368	.7548	.0250
120.5337	.3024	.2740	.7509	.3368	.7548	.0224
134.2294	.3024	.2740	.7513	.3368	.7548	.0202
149.8385	.3024	.2739	.7517	.3368	.7548	.0181
167.2363	.3023	.2739	.7520	.3368	.7549	.0162
200.4346	.3023	.2739	.7525	.3367	.7549	.0136

MACH NO = 15.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

		INVISCID AERODYNAMIC COEFFICIENTS				
L/RN	CN	CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1245	.9648	1.5198	-.1820	1.1325	1.0642
.8990	.1380	.8501	1.1501	-.0657	1.0478	.9733
1.1572	.1484	.7537	.9667	.0172	.9875	.8917
1.4746	.1573	.6615	.8545	.0867	.9369	.8085
1.7729	.1641	.5940	.8004	.1315	.9043	.7432
2.0593	.1707	.5424	.7734	.1609	.8828	.6898
2.3469	.1782	.5008	.7610	.1804	.8686	.6433
2.6013	.1854	.4707	.7571	.1918	.8604	.6072
2.8453	.1927	.4466	.7567	.1994	.8548	.5761
3.0963	.2004	.4258	.7580	.2050	.8507	.5473
3.3250	.2075	.4096	.7600	.2089	.8480	.5234
3.5507	.2144	.3959	.7622	.2119	.8457	.5019
3.7911	.2215	.3833	.7643	.2148	.8436	.4808
4.5035	.2405	.3549	.7679	.2234	.8374	.4275
5.2760	.2566	.3341	.7679	.2337	.8299	.3816
6.1478	.2698	.3178	.7651	.2457	.8211	.3404
7.0902	.2799	.3054	.7613	.2580	.8122	.3048
8.1309	.2876	.2957	.7573	.2696	.8037	.2733
9.3098	.2937	.2880	.7536	.2806	.7958	.2446
10.5922	.2981	.2822	.7504	.2901	.7888	.2195
12.0183	.3013	.2778	.7478	.2987	.7826	.1971
13.6416	.3034	.2744	.7455	.3064	.7769	.1765
15.4119	.3048	.2719	.7438	.3129	.7722	.1585
17.3834	.3055	.2700	.7425	.3185	.7682	.1423
19.5798	.3058	.2685	.7417	.3230	.7648	.1278
22.0831	.3059	.2674	.7414	.3268	.7621	.1144
24.8150	.3057	.2666	.7414	.3297	.7600	.1028
27.8574	.3055	.2659	.7417	.3319	.7584	.0923
31.3237	.3053	.2653	.7423	.3335	.7572	.0826
35.1069	.3051	.2649	.7430	.3346	.7564	.0742
39.3211	.3049	.2645	.7439	.3354	.7558	.0666
44.1223	.3048	.2643	.7448	.3360	.7554	.0597
49.3611	.3047	.2640	.7456	.3363	.7552	.0536
55.1939	.3046	.2638	.7465	.3365	.7550	.0481
61.8373	.3045	.2636	.7473	.3367	.7549	.0431
69.0854	.3044	.2635	.7481	.3367	.7549	.0387
77.1562	.3044	.2634	.7487	.3368	.7549	.0347
86.1433	.3044	.2633	.7494	.3368	.7549	.0312
96.3805	.3043	.2632	.7500	.3368	.7549	.0279
107.5503	.3043	.2631	.7505	.3367	.7549	.0251
119.9883	.3043	.2631	.7509	.3367	.7549	.0225
134.1564	.3043	.2630	.7514	.3367	.7549	.0202
149.6152	.3042	.2630	.7517	.3367	.7549	.0181
166.8294	.3042	.2630	.7521	.3367	.7549	.0163
200.1299	.3042	.2629	.7526	.3367	.7549	.0136

MACH NO = 20.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1244	.9621	1.5198	-.1820	1.1325	1.0642
.8974	.1377	.8481	1.1516	-.0662	1.0482	.9739
1.1732	.1484	.7457	.9584	.0216	.9842	.8871
1.4690	.1565	.6600	.8552	.0861	.9373	.8098
1.7840	.1635	.5888	.7982	.1334	.9029	.7410
2.0844	.1704	.5353	.7711	.1635	.8810	.6855
2.3486	.1772	.4974	.7604	.1809	.8683	.6431
2.6143	.1848	.4661	.7567	.1925	.8599	.6054
2.8680	.1924	.4413	.7567	.2001	.8544	.5734
3.0969	.1995	.4224	.7581	.2050	.8508	.5472
3.3341	.2069	.4057	.7604	.2087	.8481	.5225
3.5533	.2137	.3925	.7626	.2115	.8460	.5016
3.7868	.2208	.3802	.7648	.2142	.8440	.4811
4.5092	.2402	.3514	.7687	.2228	.8378	.4271
5.2776	.2563	.3307	.7686	.2330	.8304	.3815
6.1442	.2696	.3145	.7658	.2451	.8216	.3405
7.1009	.2798	.3018	.7617	.2576	.8125	.3044
8.1351	.2876	.2921	.7577	.2692	.8040	.2731
9.3046	.2937	.2844	.7540	.2800	.7961	.2447
10.6049	.2983	.2785	.7508	.2898	.7891	.2193
12.0212	.3015	.2741	.7481	.2983	.7829	.1970
13.6312	.3038	.2707	.7459	.3059	.7773	.1766
15.4259	.3052	.2681	.7441	.3126	.7725	.1584
17.3835	.3060	.2662	.7428	.3181	.7685	.1423
19.6118	.3064	.2647	.7420	.3228	.7651	.1276
22.0981	.3064	.2636	.7416	.3265	.7623	.1144
24.8105	.3063	.2627	.7416	.3294	.7602	.1028
27.8964	.3062	.2620	.7419	.3317	.7586	.0921
31.3383	.3060	.2615	.7425	.3333	.7574	.0826
35.0936	.3058	.2610	.7432	.3345	.7565	.0742
39.3672	.3056	.2607	.7440	.3353	.7559	.0665
44.1341	.3055	.2604	.7449	.3359	.7555	.0597
49.3332	.3054	.2601	.7457	.3362	.7552	.0536
55.2472	.3053	.2599	.7465	.3365	.7551	.0481
61.8415	.3052	.2597	.7474	.3366	.7550	.0431
69.0333	.3052	.2596	.7481	.3367	.7549	.0387
77.2149	.3051	.2595	.7488	.3367	.7549	.0347
86.3388	.3051	.2594	.7494	.3367	.7549	.0311
96.2903	.3051	.2593	.7500	.3367	.7549	.0280
107.6116	.3051	.2592	.7505	.3367	.7549	.0251
120.2368	.3051	.2592	.7510	.3367	.7549	.0225
134.0071	.3050	.2591	.7514	.3367	.7549	.0202
149.6728	.3050	.2591	.7518	.3367	.7549	.0181
167.1430	.3050	.2590	.7521	.3367	.7549	.0163
200.1423	.3050	.2590	.7526	.3366	.7549	.0136

MACH NO = 25.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID CA	AERODYNAMIC COEFFICIENTS XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1244	.9611	1.5198	-.1820	1.1325	1.0642
.8966	.1376	.8474	1.1523	-.0665	1.0484	.9741
1.1716	.1482	.7451	.9589	.0213	.9845	.8876
1.4862	.1566	.6545	.8506	.0894	.9349	.8057
1.7800	.1631	.5883	.7983	.1332	.9031	.7418
2.0788	.1698	.5349	.7710	.1634	.8811	.6864
2.3582	.1770	.4948	.7599	.1817	.8678	.6416
2.6210	.1846	.4640	.7565	.1929	.8596	.6045
2.8563	.1916	.4409	.7565	.1999	.8545	.5748
3.0977	.1991	.4209	.7581	.2050	.8508	.5471
3.3317	.2064	.4044	.7605	.2086	.8482	.5228
3.5479	.2132	.3913	.7628	.2113	.8462	.5021
3.7781	.2202	.3792	.7650	.2139	.8443	.4819
4.5204	.2402	.3496	.7690	.2226	.8380	.4263
5.3132	.2568	.3285	.7689	.2331	.8303	.3796
6.1904	.2701	.3122	.7658	.2454	.8214	.3386
7.1799	.2805	.2994	.7616	.2583	.8120	.3018
8.2532	.2883	.2896	.7575	.2702	.8033	.2700
9.4695	.2944	.2819	.7537	.2812	.7953	.2411
10.7975	.2989	.2761	.7506	.2908	.7883	.2160
12.2777	.3021	.2717	.7479	.2994	.7821	.1935
13.9640	.3043	.2684	.7457	.3071	.7765	.1729
15.8095	.3056	.2659	.7440	.3136	.7717	.1549
17.9141	.3064	.2640	.7427	.3191	.7677	.1385
20.2198	.3067	.2626	.7420	.3236	.7644	.1241
22.7941	.3067	.2616	.7417	.3272	.7618	.1112
25.7294	.3066	.2607	.7418	.3300	.7598	.0994
28.9432	.3064	.2600	.7422	.3321	.7582	.0890
32.6080	.3063	.2595	.7428	.3336	.7571	.0796
36.6220	.3061	.2591	.7435	.3347	.7563	.0713
41.1029	.3059	.2587	.7444	.3355	.7558	.0639
46.2120	.3058	.2584	.7452	.3360	.7554	.0571
51.8054	.3057	.2582	.7461	.3363	.7552	.0511
58.0466	.3056	.2580	.7469	.3365	.7551	.0458
65.1609	.3056	.2578	.7477	.3366	.7550	.0410
72.9499	.3055	.2577	.7485	.3366	.7549	.0367
81.8298	.3055	.2576	.7492	.3367	.7549	.0328
91.5421	.3055	.2575	.7498	.3367	.7549	.0294
102.1622	.3055	.2574	.7503	.3367	.7549	.0264
113.9349	.3055	.2573	.7508	.3367	.7549	.0237
126.4563	.3054	.2573	.7512	.3367	.7549	.0214
140.3001	.3054	.2572	.7516	.3367	.7549	.0193
154.9647	.3054	.2572	.7519	.3367	.7549	.0175
170.7560	.3054	.2572	.7522	.3366	.7549	.0159
200.1400	.3054	.2571	.7526	.3366	.7549	.0136

MACH NO = 30.00 CONE ANGLE = 20.00 ANGLE OF ATTACK = 10.00

L/RN	CN	INVISCID AERODYNAMIC COEFFICIENTS CA	XCP/L	YCP/D	XVCP/LV	RN/RB
.6580	.1244	.9604	1.5198	-.1820	1.1325	1.0642
.8962	.1375	.8469	1.1527	-.0667	1.0485	.9743
1.1707	.1481	.7447	.9593	.0212	.9846	.8879
1.4847	.1564	.6541	.8508	.0803	.9350	.8061
1.7969	.1632	.5842	.7960	.1354	.9015	.7384
2.0937	.1699	.5317	.7700	.1647	.8801	.6839
2.3541	.1767	.4945	.7598	.1816	.8678	.6422
2.6157	.1841	.4638	.7565	.1928	.8597	.6052
2.8652	.1916	.4393	.7565	.2001	.8543	.5737
3.1046	.1991	.4196	.7582	.2051	.8507	.5464
3.3367	.2064	.4033	.7606	.2086	.8482	.5223
3.5512	.2131	.3903	.7629	.2112	.8462	.5018
3.7796	.2201	.3783	.7652	.2138	.8444	.4817
4.5316	.2404	.3484	.7693	.2225	.8380	.4256
5.3704	.2578	.3264	.7689	.2337	.8299	.3767
6.2848	.2712	.3099	.7656	.2466	.8205	.3347
7.2974	.2814	.2973	.7612	.2596	.8110	.2980
8.4210	.2893	.2875	.7570	.2717	.8022	.2656
9.6724	.2952	.2799	.7533	.2827	.7942	.2369
11.0704	.2996	.2742	.7501	.2924	.7871	.2114
12.6669	.3028	.2699	.7474	.3012	.7807	.1883
14.4214	.3048	.2667	.7453	.3087	.7753	.1681
16.3860	.3060	.2644	.7437	.3151	.7706	.1501
18.5873	.3066	.2626	.7425	.3205	.7667	.1340
21.0544	.3068	.2613	.7419	.3249	.7635	.1196
23.8187	.3068	.2602	.7418	.3283	.7611	.1067
26.9148	.3067	.2594	.7420	.3308	.7592	.0953
30.4549	.3066	.2588	.7425	.3328	.7578	.0849
34.3490	.3064	.2583	.7432	.3341	.7568	.0757
38.7121	.3062	.2579	.7440	.3351	.7561	.0676
43.6001	.3061	.2576	.7448	.3357	.7556	.0604
49.0748	.3059	.2573	.7457	.3361	.7553	.0539
55.2057	.3059	.2571	.7466	.3364	.7551	.0481
62.0712	.3058	.2569	.7474	.3365	.7550	.0429
69.9198	.3058	.2568	.7482	.3366	.7550	.0382
78.4711	.3057	.2566	.7489	.3366	.7549	.0342
87.6922	.3057	.2565	.7496	.3367	.7549	.0307
97.6141	.3057	.2564	.7501	.3367	.7549	.0276
108.2800	.3057	.2564	.7506	.3367	.7549	.0249
119.7245	.3057	.2563	.7510	.3367	.7549	.0226
132.2191	.3057	.2563	.7514	.3367	.7549	.0205
145.2926	.3057	.2562	.7517	.3366	.7549	.0187
159.1691	.3056	.2562	.7520	.3366	.7549	.0171
173.8199	.3056	.2561	.7522	.3366	.7550	.0156
200.1149	.3056	.2561	.7526	.3366	.7550	.0136

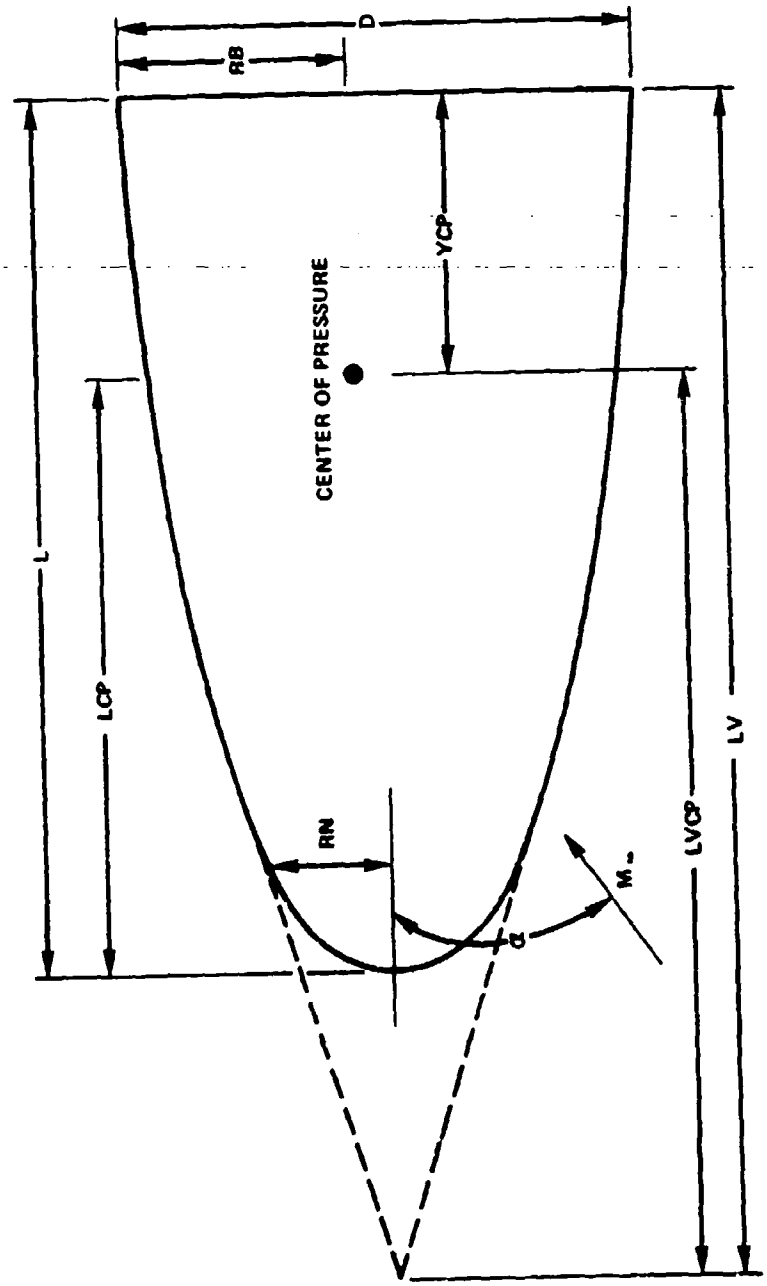

FIG. 8 DEFINITION OF TERMS USED IN AERODYNAMIC TABLES

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