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# Explosive Equivalence of Hypergolic Propellants

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Prepared for:

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The purpose of this report is to document the explosive equivalence of $N_2O_4$/A-50 and other hypergolic energetic liquid bipropellants for inclusion in DESR 6055.09 (Ref. 1). This analysis is similar to the recent report on hydrocarbon propellants that can be found in Aerospace TR-2019-00959 (Ref. 2). The focus of this report will be on revisions to $N_2O_4$/A-50 but other combinations [$N_2O_4$/$N_2H_4$, $N_2O_4$/UDMH, IRFNA/UDMH, $N_2O_4$/MMH, pentaborane (PB), and chlorine trifluoride/pentafluoride (CTF/CPF) combinations] will be addressed. Based on findings during this latest review additional recommendations for revisions to DESR 6055.09 are included. This revision is intended to apply to DoD facilities siting and construction for operations involving these energetic liquids. It does not govern the storage or handling of energetic liquids for uses other than in launch vehicles, space vehicles, rockets, missiles, and associated static test and processing installations. The yield criteria will envelope test/processing facilities and launch facilities for the pre-launch/pre-ignition state, on-pad engine firing conditions, and post-launch hazards relative to near-pad incidents. Additional assessments are included which apply to other operational facilities and conditions.

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## Abstract

This report documents the explosive equivalence of $N_2O_4$/A-50 and other hypergolic energetic liquid bipropellants for inclusion in Defense Explosives Safety Regulation (DESR) 6055.09 [1]. This analysis is similar to the analysis in the recent report on hydrocarbon propellants (Aerospace Report Number TR-2019-00959 [2]). This report focuses on revisions to $N_2O_4$/A-50, but other combinations ($N_2O_4$/$N_2H_4$, $N_2O_4$/UDMH, IRFNA/UDMH, $N_2O_4$/MMH, pentaborane, and chlorine trifluoride/pentafluoride combinations) will be addressed. Based on findings during this latest review, additional recommendations for revisions to DESR 6055.09 are included.

This revision is intended to apply to Department of Defense facilities siting and construction for operations involving these energetic liquids. It does not govern the storage or handling of energetic liquids for uses other than in launch vehicles, space vehicles, rockets, missiles, and associated static test and processing installations. The yield criteria will envelope test/processing facilities and launch facilities for the pre-launch/pre-ignition state, on-pad engine firing conditions, and post-launch hazards relative to near-pad incidents. Additional assessments are included, which apply to other operational facilities and conditions.

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# 1. Introduction

This report is follow-on to preliminary observations made to the DDESB Liquid Propellant Working Group (LPWG) at the time the $LO_2/LH_2$ cryogenic analysis was performed, as noted in Reference [3]. This prior effort was reviewed in detail in [2] and will not be repeated here. An additional assessment of hypergolic propellant hazards was conducted in 2008 through 2009 [4]. That assessment was specific to hazardous processing of space vehicles, but additional data were generated during that effort, which expanded the database used in the 1998 timeframe. Appendix A contains backup discussions on the hypergolic bipropellants related to prediction methods, blast asymmetry, impact sensitivity, and past programs.

During the period of LPWG deliberations, the preliminary observations for nitrogen tetroxide ($N_2O_4$)/ Aerozine-50 (A-50) propellants were presented and plans for updating those explosive criteria were put in place. This report presents the follow-on assessments that were deferred at that time. In addition, an addendum is being prepared to update [6] to document the findings from these most recent efforts.

This report details these latest activities and recommends that an adjustment to the explosive siting criteria be made accordingly. In this report, the hypergolic propellants are subcategorized as the hydrazines (typically combined with a nitric acid or nitrogen tetroxide), boranes ($B_2H_6$, $B_5H_9$, and $B_{10}H_{14}$) and halogenated fluorides ($ClF_3$ and $ClF_5$). Other hypergolic bipropellant combinations have been proposed, used, and tested in the past (e.g., anilines, amines, furfuryl alcohol, etc.) but are no longer used operationally in the U.S., so they should be addressed separately, as needed.

The data used in the initial release of Department of Defense (DOD) 6055.9-STD [7] accounted for the difference in yield noted during early testing of the $N_2O_4$/A-50 propellant combination for missile programs and represented the conclusion as adjusted for propellant mass, i.e., 5% trinitrotoluene (TNT) for test stands and 10% TNT for launch pads. In addition to revising the explosive standard for $LO_2/LH_2$, the LPWG entertained a potential change to the $N_2O_4$/A-50 standards based on updated test and missile failure data. This test and failure data were reanalyzed and presented to the LPWG, as documented in [5]. For this current report, additional test and failure data were uncovered after the preliminary assessment was prepared and are included herein. In addition to the $N_2O_4$/A-50 data, testing with $N_2O_4$/hydrazine ($N_2H_4$), $N_2O_4$/monomethylhydrazine (MMH), $N_2O_4$/unsymmetrical dimethyl hydrazine (UDMH), $B_5H_9/N_2H_4$, $N_2O_4/B_5H_9$, and halogenated fluoride combinations $ClF_3/B_5H_9$ and $ClF_3/N_2H_4$ was uncovered.

The following recommendation options are consistent with the decisions made by the DDESB relative to the current propellant yield curves. Use of a yield curve incorporates the finding that liquid bipropellant explosive yields vary with mass and cannot be accurately approximated with a constant TNT percent equivalent explosive weight (EEW) as currently employed. This report includes recommendations for $N_2O_4$/A-50, $N_2O_4/N_2H_4$, $N_2O_4$/MMH, $N_2O_4$/UDMH, $B_5H_9/N_2H_4$, $N_2O_4/B_5H_9$, $ClF_3/N_2H_4$, and $ClF_5/N_2H_4$. It is acknowledged that these hypergolic combinations are no longer used on launch vehicles or missiles. However, the recommendations are made for consistency, to differentiate the various combinations, and for the historical record. In the case of the $N_2H_4$, MMH, and UDMH cases, very few tests were conducted to arrive at independent values, but observations can be made for them by comparing similar tests conducted with similar commodities. The pentaborane combinations $B_5H_9/N_2H_4$ and $B_5H_9/N_2O_4$ were found to exhibit the greatest yields, while in the case of the chlorine trifluoride/pentafluoride (CTF/CPF) combinations, the explosive yields were extremely small due to their inherent reactivity.

# 2. Recommendation Summary

## 2.1 $N_2O_4$/A-50

The recommended change to the $N_2O_4$/A-50 liquid propellant explosive equivalence is to replace the current 5% (static test facility) and 10% (launch facility) TNT values in [1], Table V5.E4.T5, with the explosive equivalence from the recommended $N_2O_4$/A-50 Yield Curves shown in Figure 1. For operations other than static test stands and launch pads (i.e., other potential explosive site (PES)) and quantities less than $W_{Pmin}$, a separate explosive yield is recommended. The following subsections describe the methods.

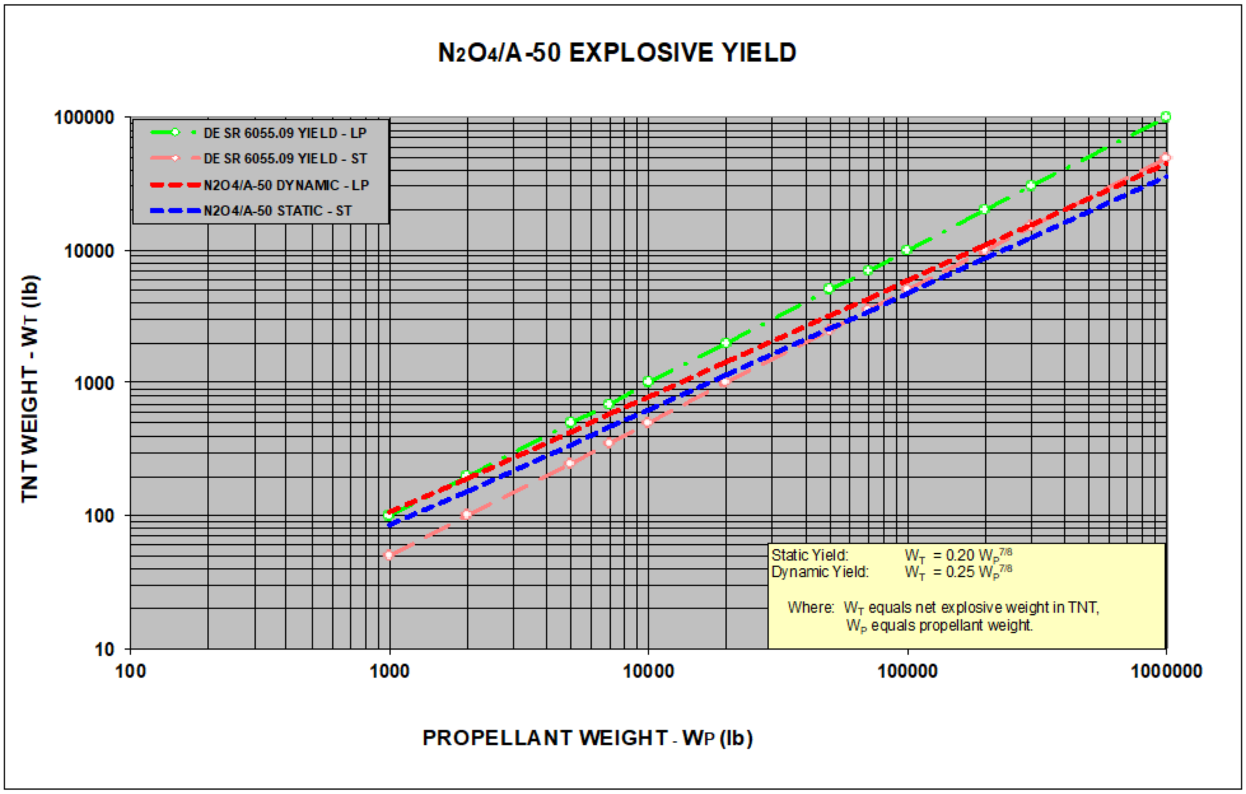


Figure 1. $N_2O_4$/A-50 proposed explosive Yield Curves.

### 2.1.1 Static Test or Processing Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities, the EEW in pounds of TNT is determined by the $N_2O_4$/A-50 Static Yield Curve shown as ST in Figure 1:

$$W_T = 0.20\ W_P^{7/8} \text{ for all } W_P \geq W_{Pmin}$$

### 2.1.2 Launch Facilities

For launch vehicles, space vehicles, or missiles in the launch configuration, determine the EEW in pounds of TNT is determined by the $N_2O_4$/A-50 Dynamic Yield Curve shown as LP in Figure 1:

$$W_T = 0.25\ W_P^{7/8} \text{ for all } W_P \geq W_{Pmin}$$

### 2.1.3 Other PESs

For operations other than those for launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

$$W_T = 14\% W_P$$

where

> $W_T$ = net explosive weight (lb) TNT, and
> $W_P$ = propellant weight (lb), as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
> $W_{Pmin}$ = 1,000 lb, and
> mixed oxides of nitrogen (MON) can be substituted for $N_2O_4$

## 2.2 $N_2O_4/N_2H_4$

The recommended change is to include $N_2O_4/N_2H_4$ as a separate liquid bipropellant combination. Current criteria only call out inhibited red fuming nitric acid (IRFNA)/UDMH explicitly as 10% (static test facility) and 10% (launch facility) TNT explosive equivalence in [1], Table V5.E4.T5, but imply 5% (static test facility) and 10% (launch facility) TNT explosive equivalence values for the $N_2O_4/N_2H_4$ combination. It is recommended that these values be replaced using the recommended $N_2O_4/N_2H_4$ Yield Curve shown in Figure 2, using the methods described in the following subsections.

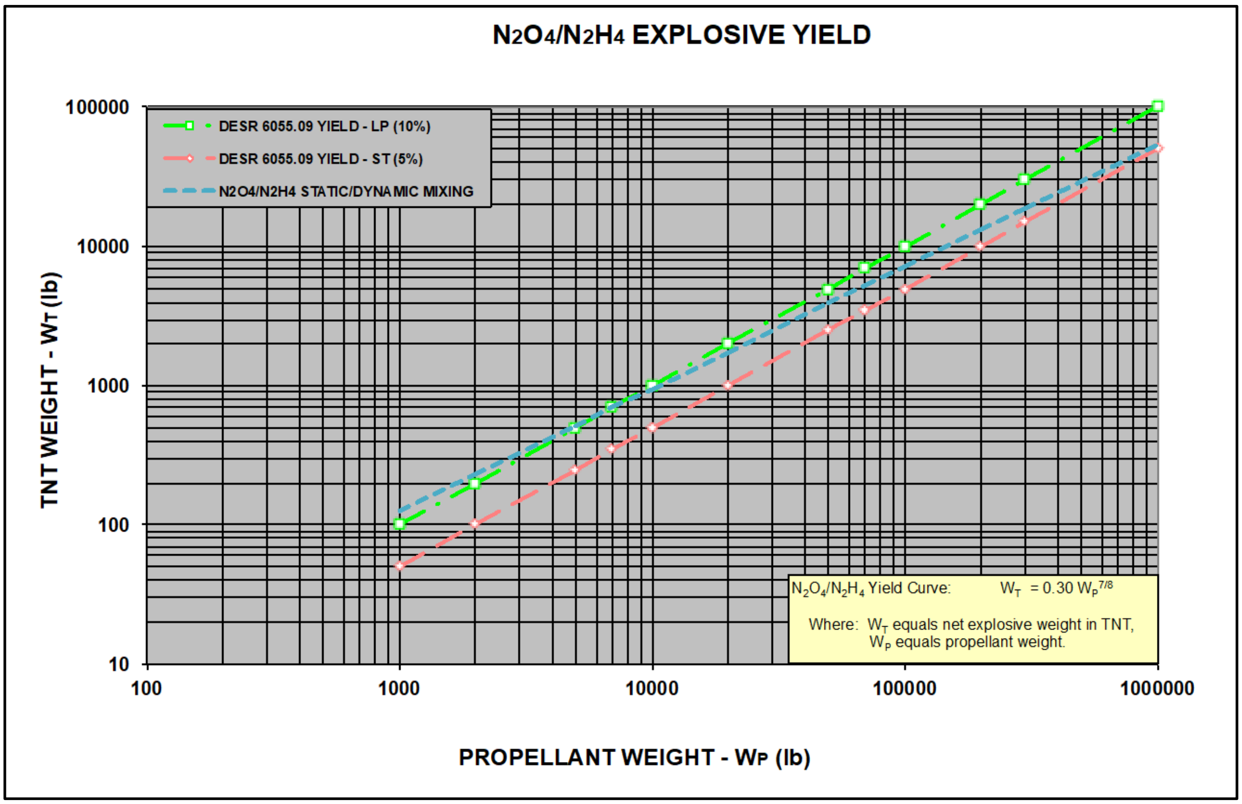


Figure 2. $N_2O_4/N_2H_4$ proposed explosive Yield Curves.

### 2.2.1 Static Test or Processing Facilities and Launch Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities and in the launch configuration, the EEW in pounds of TNT is determined by the $N_2O_4/N_2H_4$ Yield Curve:

$$N_2O_4/N_2H_4: \quad W_T = 0.30\ W_P^{7/8} \text{ for all } W_P \geq W_{Pmin}$$

### 2.2.2 Other PESs

For operations other than those for launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

$$N_2O_4/N_2H_4: \quad W_T = 18\% W_P$$

where

$W_T$ = net explosive weight (lb) TNT, and
$W_P$ = propellant weight (lb), as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
$W_{Pmin}$ = 1,000 lb, and
MON can be substituted for $N_2O_4$

## 2.3 $N_2O_4$/MMH

The recommended change is to include $N_2O_4$/MMH as a separate liquid bipropellant combination. Current criteria only call out IRFNA/UDMH explicitly as 10% (static test facility) and 10% (launch facility) TNT explosive equivalence in [1], Table V5.E4.T5, but imply 5% (static test facility) and 10% (launch facility) TNT explosive equivalence values for the $N_2O_4$/MMH combination. It is recommended that these values be replaced using the recommended $N_2O_4$/MMH Yield Curve shown in Figure 3, using the methods described in the following subsections.

### 2.3.1 Static Test or Processing Facilities and Launch Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities and in the launch configuration, the EEW in pounds of TNT is determined by the $N_2O_4$/MMH Yield Curve:

$$N_2O_4/\text{MMH}: \quad W_T = 0.28\ W_P^{7/8} \text{ for all } W_P \geq W_{Pmin}$$

### 2.3.2 Other PESs

For operations other than those for launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

$$N_2O_4/\text{MMH}: \quad W_T = 16\% W_P$$

where

$W_T$ = net explosive weight (lb) TNT, and
$W_P$ = propellant weight (lb) as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
$W_{Pmin}$ = 1,000 lb, and

MON can be substituted for $N_2O_4$

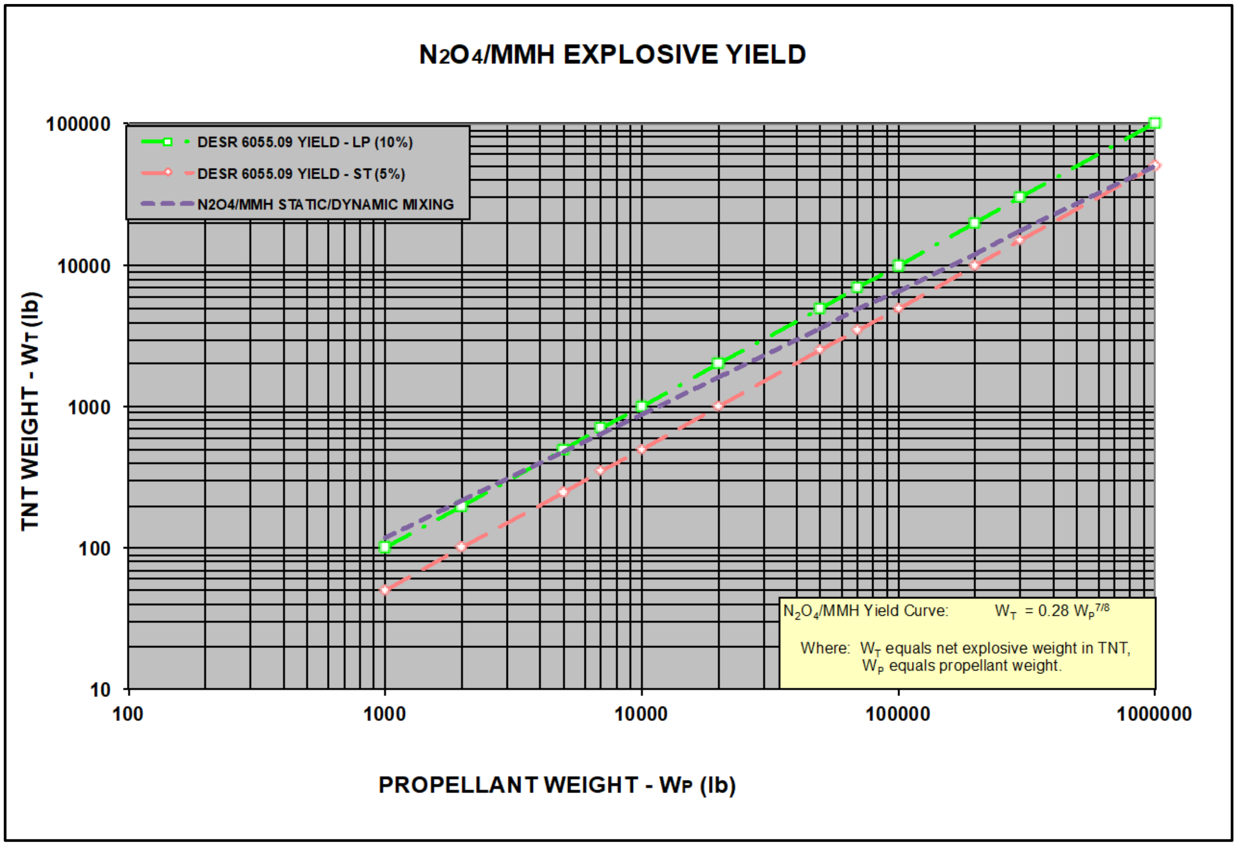


Figure 3. $N_2O_4$/MMH proposed explosive Yield Curve.

## 2.4 $N_2O_4$/UDMH and IRFNA/UDMH

The recommended change is to include $N_2O_4$/UDMH and IRFNA/UDMH as separate liquid bipropellant combinations. Current criteria only call out IRFNA/UDMH explicitly as 10% (static test facility) and 10% (launch facility) TNT explosive equivalence in [1], Table V5.E4.T5, but imply 5% (static test facility) and 10% (launch facility) TNT explosive equivalence values for the $N_2O_4$/UDMH combination. It is recommended that these values be replaced using the recommended yield curves shown in Figure 4 using the methods described in the following subsections.

### 2.4.1 Static Test or Processing Facilities and Launch Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities and in the launch configuration, the EEW in pounds of TNT is determined by the yield curves:

$N_2O_4$/UDMH: $W_T = 0.19\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$

IRFNA/UDMH: $W_T = 0.26\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$

### 2.4.2 Other PESs

For operations other than those for launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

$N_2O_4$/UDMH: $W_T = 12\% W_P$
IRFNA/UDMH: $W_T = 15\% W_P$

where

$W_T$ = net explosive weight (lb) TNT, and
$W_P$ = propellant weight (lb) as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
$W_{Pmin}$ = 1,000 lb, and
MON can be substituted for $N_2O_4$

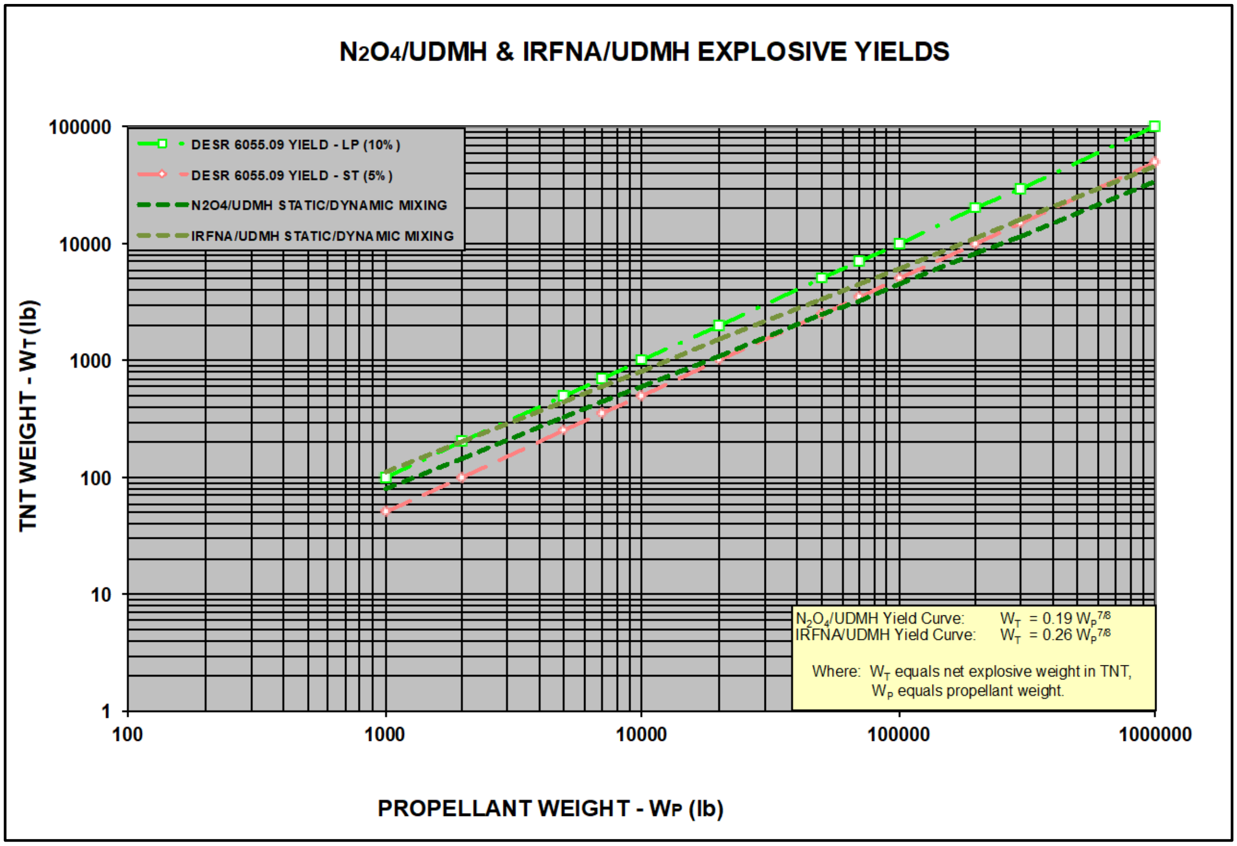


Figure 4. $N_2O_4$/UDMH and IRFNA/UDMH proposed explosive Yield Curves.

## 2.5 Pentaborane (PB) Hypergolic Combinations (Tentative Option)

Although no longer available as a rocket propellant, an option to consider is maintaining a database in DESR 6055.09 [1] of alternate propellant combinations including the pentaborane ($B_5H_9$) hypergolic combinations ($B_5H_9/N_2H_4$ and $N_2O_4/B_5H_9$) as separate liquid bipropellant combination entries. Currently these combinations are not included; however, at one time, the tables in [1] included 10% (static test facility) and 20% (launch facility) TNT explosive equivalence values for $B_5H_9/N_2H_4$ and 60% (static test or launch facility) for $N_2O_4/B_5H_9$. The explicit values for each combination from this assessment are the PB Yield Curves shown in Figure 5,using the methods described in the following subsections.

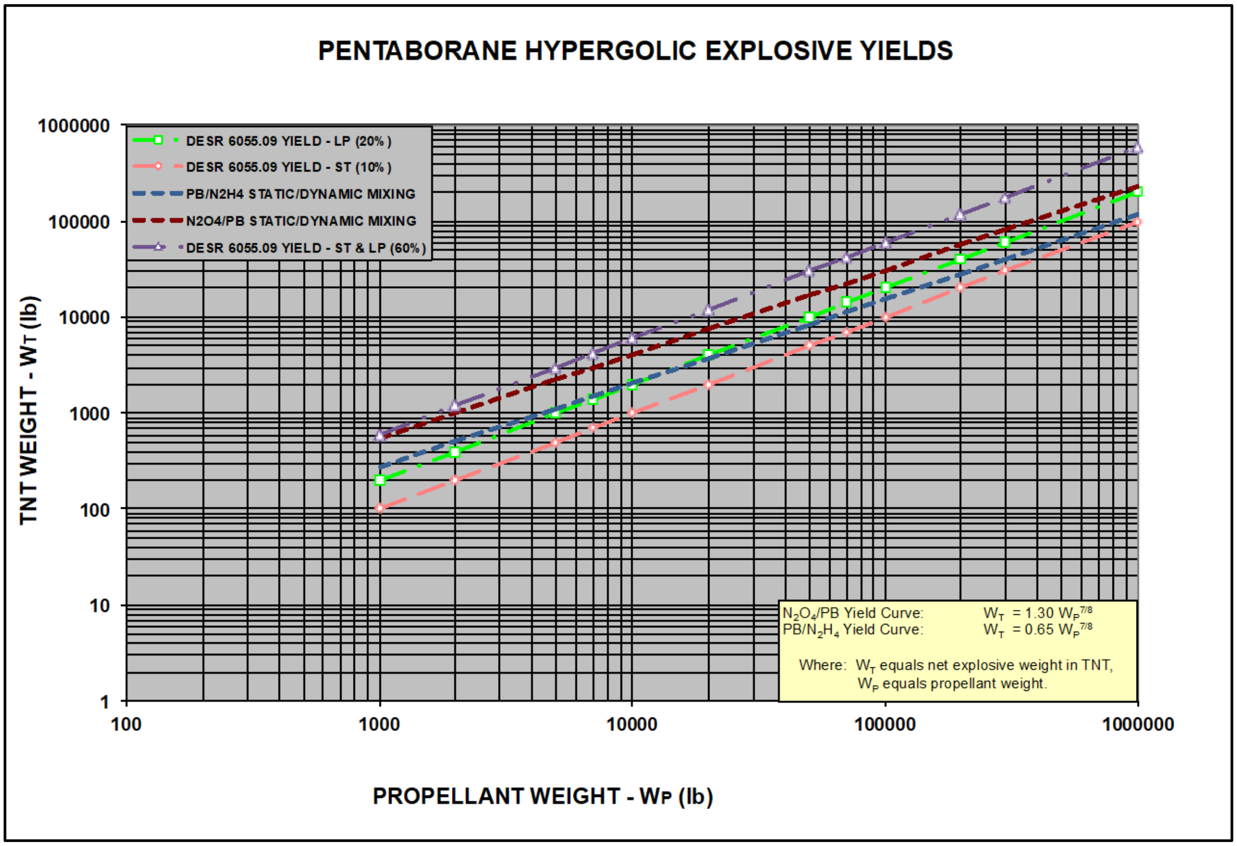


Figure 5. $PB/N_2H_4$ and $N_2O_4/PB$ proposed explosive Yield Curves.

### 2.5.1 Static Test or Processing Facilities and Launch Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities and in the launch configuration, the EEW in pounds of TNT is determined by the PB Yield Curves:

$B_5H_9/N_2H_4$: $W_T = 0.65\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$
$N_2O_4/B_5H_9$: $W_T = 1.30\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$

### 2.5.2 Other PESs

For operations other than those for launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

$B_5H_9/N_2H_4$: $W_T = 40\%W_P$
$N_2O_4/B_5H_9$: $W_T = 75\%W_P$

where

$W_T$ = net explosive weight (lb) TNT, and
$W_P$ = propellant weight (lb) as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
$W_{Pmin}$ = 1,000 lb, and
MON can be substituted for $N_2O_4$

## 2.6 CTF/CPF Hypergolic Combinations (Tentative Option)

Another option to consider is including CTF/CPF hypergolic combinations (chlorine trifluoride ($ClF_3$)/$N_2H_4$, $ClF_5$/$N_2H_4$, and $ClF_3$/PB) as separate liquid bipropellant combinations in a DESR 6055.09 [1] database of alternate propellant combinations. Currently these combinations are not included; however, at one time, the tables in [1] included 10% and 20% TNT explosive equivalence values for these combinations. The explicit values for each combination from this assessment are the CTF/$N_2H_4$ Yield Curve shown in Figure 6, using the methods described in the following subsections.

### 2.6.1 Static Test or Processing Facilities and Launch Facilities

For launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities and in the launch configuration, the EEW in pounds of TNT is determined by the CTF/$N_2H_4$ Yield Curve:

| | |
|---|---|
| $ClF_3/N_2H_4$: | $W_T = 0.003\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$<br>or use individual commodity siting criteria in [1], Table V5.E4.T3 |
| $ClF_3/B_5H_9$: | Use individual commodity siting criteria in [1], Table V5.E4.T3 |

### 2.6.2 Other PESs

For operations other than launch vehicles, space vehicles, missiles, stages, engines, or components in static test or processing facilities; launch vehicles or missiles in the launch configuration; or quantities less than $W_{Pmin}$, the EEW in pounds of TNT is determined by the following:

| | |
|---|---|
| $ClF_3/N_2H_4$: | $W_T = 1\% W_P$<br>or use individual commodity siting criteria in [1], Table V5.E4.T3 |
| $ClF_3/B_5H_9$: | Use individual commodity siting criteria in [1], Table V5.E4.T3 |

where

$W_T$ = net explosive weight (lb) TNT, and
$W_P$ = propellant weight (lb) as defined in [1], V5.E4.5.6.1 and V5.E4.5.6.2, and
$W_{Pmin}$ = 1,000 lb

Explosive data related to $ClF_5/N_2H_4$ are somewhat obscure due to the classification of this combination during its early testing. Although the performance of CPF is known to be greater than CTF, their hypergolic reactions are similar; therefore, the $ClF_3/N_2H_4$ explosive yields noted above are likely acceptable as interim values for $ClF_5/N_2H_4$. However, a comprehensive test program is needed before an explosive yield can be formally recommended.

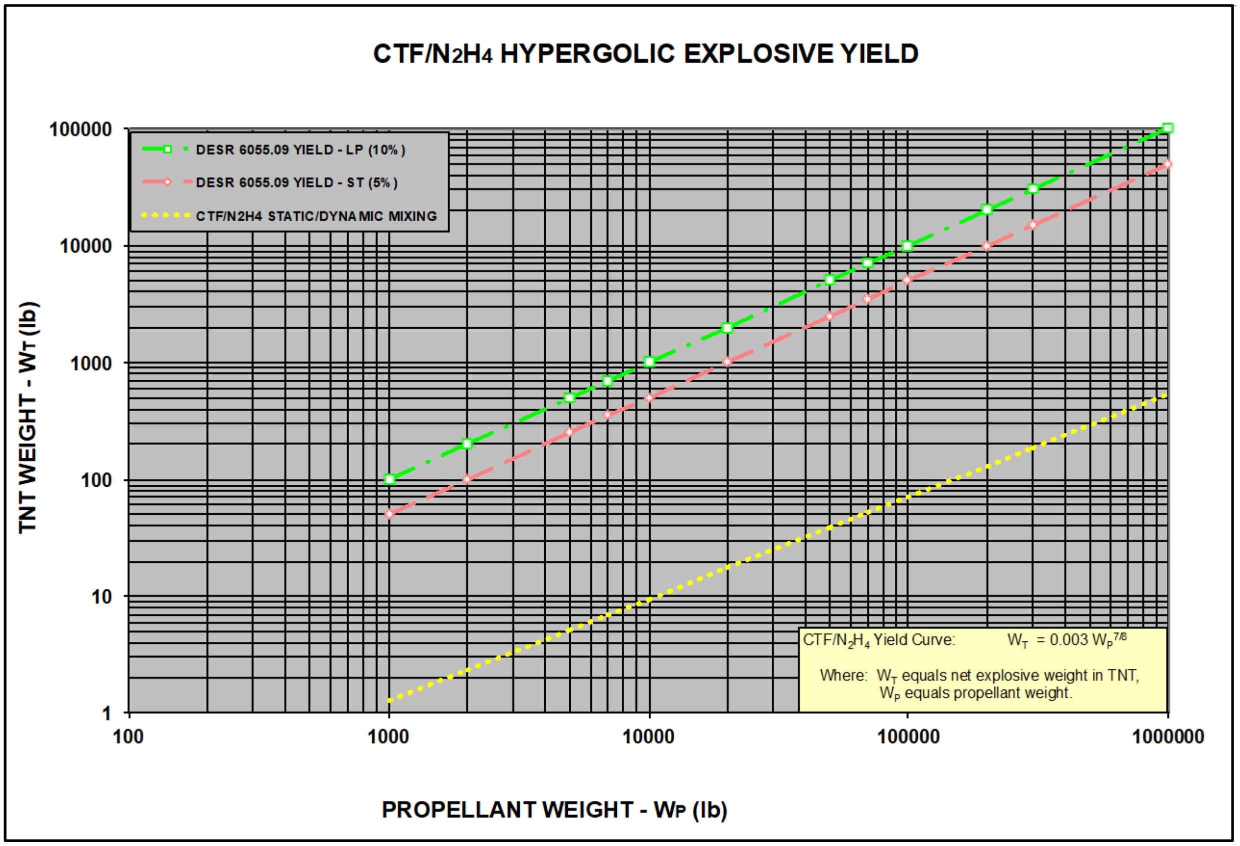


Figure 6. CTF/$N_2H_4$ proposed explosive Yield Curve.

## 2.7 Application

Use of the $N_2O_4$/A-50 and IRFNA/UDMH propellant combinations dates to the early days of the missile and launch vehicle programs. With the recent retirement of the Delta II, it is highly unlikely that either will ever be used again in large quantities. However, they remain as options to use on test programs or space vehicles along with $N_2O_4/N_2H_4$ and $N_2O_4$/MMH. Current uses of CTF and CPF are unknown, but $ClF_3/N_2H_4$ and $ClF_5/N_2H_4$ remain available as bipropellants in missile applications. In fact, $ClF_5/N_2H_4$ remains an attractive hypergolic combination due to its performance. Research into use of PB as a propellant has been discontinued; however, although the PB and CTF/CPF values are considered appropriate, due to the nature of the limited data used to establish the above yield values it is recommended that this information be retained in an appropriate database and a note added that these values could be used in lieu of a new comprehensive explosive test program, which would be required to establish an appropriate EEW pending any plans for reuse.

Other hypergolic combinations have recently been proposed, such as WFNA/turpentine and WFNA/furfuryl alcohol, but no operational systems have been developed. Use of such combinations in launch vehicle quantities must be analyzed using the appropriate data in this report or undergo specific explosive testing. In addition, other hypergolic propellants remain in use for engine starting circuits (e.g., furfuryl alcohol); however, such usage is in small quantities that are not governed by the requirements of DESR 6055.09 [1].

# 3. Background

Per [6], the current launch pad explosive factor of 10% for $N_2O_4$/A-50 found in [7] (and prior versions) was based on hazard analysis estimates in advance of obtaining definitive test results. The 10% factor for $N_2O_4$/A-50 appears initially in [9]. This hazard analysis was performed for a large theoretical vehicle using $N_2O_4$/A-50 and $LO_2$/RP-1. The 10% value used was conservatively applied to the entire vehicle and was based principally on $LO_2$/RP-l information. The derivation of the 5% explosive factor used for static test stands, however, has not been found.

In the very early days of the missile and sounding rocket programs, nitric acid (red fuming nitric acid (RFNA), white fuming nitric acid (WFNA), IRFNA, etc.) and UDMH were popular hypergolic propellant combinations. Explosive testing of these combinations appears limited to specialized testing by the Bureau of Mines during the 1940s and 1950s. Then in the early 1960s, attention was placed on the new storable hypergolic propellants planned for the Titan missile. This testing and the results are discussed in detail in [6]. The primary testing was done by Rocketdyne and involved the hypergolic combinations $N_2O_4/N_2H_4$, $ClF_3/N_2H_4$, $ClF_3/B_5H_9$, $B_5H_9/N_2H_4$, $N_2O_4/B_5H_9$, and $N_2O_4$/A-50. This test program is discussed below.

Other than Aerojet-General testing for NASA's Gemini and Apollo test programs, the only other significant testing done with $N_2O_4$/A-50 was performed by Project PYRO. It is evident that the PYRO data were not used to re-evaluate the $N_2O_4$/A-50 criteria, and it is unknown whether any additional discussions took place in this time frame. This report is the first known effort to re-assess the explosive potential and appropriate yield values for the various hypergolic combinations since the mid-1960s.

## 3.1 Propellant Test Programs Overview

Table 1 shows a matrix of liquid propellant test programs that have been reviewed [10]–[57]. The table gives the experimenter, program, date, propellant combinations (cryogenic $LO_2/LH_2$, various hypergolic pairs, $LO_2$/hydrocarbons (HCs), $LO_2/LH_2$/RP-1 tri-propellants, and $H_2O_2$/hydrocarbon), and mixing method.

Table 1. Liquid Propellant Test Programs

| Agency/Test Program | Propellants | | | | |
|---|---|---|---|---|---|
| | $LO_2/LH_2$ | Hyper | $LO_2$/HC | Tri | $H_2O_2$/HC |
| NAVY/PUSHOVER (1948) | | | | | |
| Vehicle toppling | | | X | | |
| NAVY/NRL TESTS (1949) | | | | | |
| Propellant mixing | | | X | | |
| BUREAU OF MINES (1949) | | | | | |
| Ballistic mortar test | | X | | | |
| Detonability test | | X | | | |
| AEROJET GENERAL (1952) | | | | | |
| TBD | | X | | | |
| ROCKETDYNE/FIELD TESTS (1954-1956) | | | | | |
| Propellant mixing | | | X | | |
| Ground spill | | | X | | |
| Impinging streams | | | X | | |

Table 1. Liquid Propellant Test Programs (continued)

| Agency/Test Program | Propellants | | | | |
|---|---|---|---|---|---|
| | $LO_2/LH_2$ | Hyper | $LO_2$/HC | Tri | $H_2O_2$/HC |
| ARMY/REDSTONE (1956) | | | | | |
| Vehicle destruct | | | X | | |
| NAVY/JUPITER (1956) | | | | | |
| Propellant mixing | | | | | |
| Ground spill | | | X | | |
| Vehicle destruct | | | X | | |
| Tank fallback | | | X | | |
| Tank fallback–water | | | X | | |
| Underwater | | | X | | |
| BROADVIEW/ATLAS (1957-1961) | | | | | |
| Ground spill | | | X | | |
| Tank rupture-ground | | | X | | |
| Tank fallback | | | X | | |
| Tank rupture–in silo | | | X | | |
| BUREAU OF MINES/NAVY (1959-1960) | | | | | |
| Propellant mixing | | X | | | |
| A. D. LITTLE/CENTAUR (1960) | | | | | |
| Gas phase | X | | | | |
| Intertank mixing | X | | X | X | |
| Ground spill | X | | X | X | |
| A. D. LITTLE/SATURN (1961) | | | | | |
| Ground spill | X | | X | X | |
| ROCKETDYNE/STORABLE (1961) | | | | | |
| Ground spill | | X | | | |
| Tank rupture | | X | | | |
| ROCKETDYNE/TITAN II (1961) | | | | | |
| Ground spill | | X | | | |
| Tank rupture–ground | | X | | | |
| Tank rupture–in silo | | X | | | |
| Propellant mixing | | X | | | |
| ATLANTIC RESEARCH/LAB TESTS (1962) | | | | | |
| Ground spill | | X | | | |
| Falling droplet | | X | | | |
| Liquid I/F injection | | X | | | |
| Gas/liquid | | X | | | |
| MARTIN MARIETTA/TITAN II (1962) | | | | | |
| Vehicle destruct | | X | | | |
| NASA MSFC/SATURN I (1963) | | | | | |
| Vehicle destruct | | | X | | |
| AEROJET/GEMINI (1963-1964) | | | | | |
| Distributed mixing* | | X | X | | |
| Intertank mixing | | X | X | | |
| Tank fallback | | X | X | | |
| URS/PYRO (1963-1968) | | | | | |
| Confined-by-missile mixing | X | X | X | | |
| Confined-by-ground mixing | X | X | X | | |
| Tank fallback | | X | | X | |
| High velocity impact | X | X | X | | |
| Vehicle destruct | | X | | | |
| Donor | | X | | | |

Table 1. Liquid Propellant Test Programs (continued)

| Agency/Test Program | Propellants | | | | |
|---|---|---|---|---|---|
| | $LO_2/LH_2$ | Hyper | $LO_2$/HC | Tri | $H_2O_2$/HC |
| AEROJET (1964) | | | | | |
| Distributed mixing* | | X | | | |
| AEROJET/APOLLO (1965) | | | | | |
| Distributed mixing* | X | X | X | | |
| NASA MSFC/SATURN (1965) | | | | | |
| Ground spill | X | | X | | |
| Intertank mixing | X | | X | | |
| Vehicle destruct | X | | X | | |
| BUREAU OF MINES/GEMINI (1966) | | | | | |
| Ballistic mortar test | | X | | | |
| Distributed mixing* | X | | | | |
| NASA WSTF/CASSINI-PRC (1994-1995) | | | | | |
| Distributed mixing* | | X | X | | X |
| NASA WSTF/HOVI (1996) | | | | | |
| Tank fallback | X | | | | |
| *Distributed mixing is also referred to as immersion mixing elsewhere. | | | | | |

## 3.2 Failure History Overview

Data were obtained and analyzed [58]–[73] from the near-pad failures of 33 liquid-propellant launch vehicles. These included the vehicle types shown in Table 2. Of the 33 failures studied, 2 were static test stand failures, 30 failed on or near the launch pad/silo, 1 occurred in flight, 6 occurred prior to engine ignition, 13 were instrumented, 7 were destruct cases, and for 4 events, explosive data have not been obtained.

Table 2. Liquid Propellant Launch Vehicle Failure History

| Vehicle | Failures | Propellants |
|---|---|---|
| ANTARES | 1 | $LO_2$/RP-1/HTPB |
| ATLAS | 5 | $LO_2$/RP-1 |
| ATLAS-CENTAUR | 1 | $LO_2$/RP-1/$LH_2$ |
| DELTA II | 1 | HTPB/$LO_2$/RP-1 |
| FALCON 9 | 1 | $LO_2$/RP-1 |
| JUPITER/JUNO | 3 | $LO_2$/RP-1 |
| LONG MARCH | 1 | $N_2O_4$/UDMH/$LO_2/LH_2$ |
| N-1 | 1 | $LO_2$/Kerosene |
| NAVAHO | 1 | $LO_2$/Ethanol/JP-4 |
| REDSTONE | 1 | $LO_2$/Ethanol |
| SATURN S-IV | 2 | $LO_2/LH_2$ |
| SPACE SHUTTLE | 1 | $LO_2/LH_2$ |
| THOR | 4 | $LO_2$/RP-1 |
| TITAN I | 3 | $LO_2$/RP-1 |
| TITAN II | 4 | $N_2O_4$/A-50 |
| TITAN 34D | 1 | PBAN/$N_2O_4$/A-50 |
| VANGUARD | 1 | $LO_2$/RP-1 |
| V-2 | 1 | $LO_2$/Ethanol |

# 4. Review of Past Hypergolic Liquid Test Projects

Following is a brief discussion of the hypergolic explosive test projects. The discussions are organized according to the experimenter.

## 4.1 Bureau of Mines (BoM)

Testing of hypergolic propellants by the BoM dates to 1949 with the investigation of the explosive properties of hydrazine. Hypergolic propellant testing was performed by the BoM Explosive Research Laboratory. After the monopropellant hydrazine tests, the BoM began testing hypergolic bipropellant combinations under contract to the Navy Bureau of Ships and the Air Force Aeronautical Systems Division from 1959 through 1964. Additional testing was performed for NASA/Manned Spacecraft Center in 1966.

### 4.1.1 Hydrazine Explosive Tests

In order to understand the hazards associated with employing $N_2H_4$ and hydrazine hydrate ($N_2H_4$-$H_2O$) as fuels, the Department of the Interior (DoI) BoM Explosives Research Branch investigated the properties and explosive hazards associated with manufacture, storage, transportation, and use [11]. Explosive hazard tests included sensitivity to impact, friction, and electrostatic discharge; ballistic mortar TNT equivalence tests; and "Mettegang recorder" detonability tests. The ballistic mortar tests (4 tests) were conducted using 10 g (0.02 lb) of propellant. The BoM reported an $N_2H_4$ TNT equivalence of 135%. When testing for detonability using 0.46 lb of propellant, no detonations were recorded (4 tests).

### 4.1.2 New Liquid Propellant

Research was conducted for the Office of Naval Research to investigate the explosion hazards associated with a variety of new liquid propellants and propellant combinations [28][30] Testing was defined in three phases: ignition, flame propagation and detonability, and burning. Hypergolic propellants tested include Report No. 7 [30] - $N_2H_4$, MMH, UDMH, MAF-1, MAF-3, methanol, and benzene. Air, $N_2O_4$, and $O_2$/air were used as oxidants. Results are to be supplied (TBS).

### 4.1.3 Flight Vehicle Hazards

Research was conducted for the Air Force Systems Command to investigate the aircraft and missile explosion hazards associated with a variety of combustible propellants and propellant combinations [31][32][36]. Results are TBS.

### 4.1.4 Damage Assessments

In 1966, following the failure of the Gemini IV Agena (IRFNA/UDMH) target vehicle engine and Apollo reaction control system (RCS) thrusters ($N_2O_4$/A-50) during altitude firing, NASA/Manned Spacecraft Center requested that the BoM's Explosives Research Center investigate the combustion characteristics of $N_2O_4$/A-50 [49]. In addition to damage assessments, the objective included determining the explosive potential of the oxidizer and fuel both individually and in combination. The BoM used their ballistic mortar test approach to determine the TNT equivalence of the propellants. The ballistic mortar test measures the energy imparted to a 36.6 lb projectile fired from a 500 lb mortar and compares the explosive energy to TNT. Claimed resolution is within ± 1% TNT equivalence. Two types of tests were conducted using 5 g (0.01 lb) of propellant: (1) individual propellants ($N_2H_4$, UDMH, MMH, $N_2O_4$) were fired in various atmospheres using a no. 8 detonator, and (2) liquid fuel and $N_2O_4$ stoichiometric combinations were explosively combined in a nitrogen atmosphere. In both cases, the liquids were contained in glass vessels and the detonator encapsulated in glass under the liquid. The kinetic energy delivered to the mortar is presumed to be proportional to the energy released in the reaction. The test apparatus was calibrated using TNT. The 5 g (0.01 lb) bipropellant tests performed (6 tests each) resulted in 90% to 187% TNT explosive yields. The $N_2O_4$/A-50 and $N_2O_4$/$N_2H_4$ tests gave similar results (average 160%), while the $N_2O_4$/MMH and $N_2O_4$/UDMH tests were on the order of 115% (approximately 70% less). However, the $N_2O_4$/A-50 values were believed to be greater than expected due to the test configuration.

## 4.2 Aerojet-General

### 4.2.1 Aerojet-General (Aerobee)

A test program was performed [16] to develop safety and design criteria for storable liquid propellants. Results are TBS.

### 4.2.2 Aerojet-General (Gemini)

In 1963, Aerojet-General conducted experiments that supported NASA/Manned Spacecraft Center hazard assessments for the Gemini program [42][43]. The tests were conducted to obtain better estimates of the blast hazards associated with using new (i.e., other than $LO_2$/RP-1) propellant combinations on manned programs. The test approach was to mix constant 300 lb total weights of fuel/oxidizer combinations at varying mixture ratios and contact area ratios. Propellants used were $N_2O_4$/A-50 for the Gemini-Titan launch vehicle and $LO_2$/RP-1 for comparison. Mixing energy was achieved by dropping oxidizer-filled glass dewars, which were immersed in an aluminum pan filled with fuel, from a height of 20 feet. The purpose of the initial test program was to quantify the $N_2O_4$/A-50 blast hazard potential compared to the $LO_2$/RP-1 propellant combination. A second test phase was conducted which simulated propellant mixing during launch vehicle booster failures. However, no conclusions were made or inferred relative to full-scale failure modes or effects.

#### 4.2.2.1 Propellant Mixing Tests

Test data were obtained for peak overpressure, positive impulse, and fireball history [42]. The overpressure impulse data were referenced to TNT equivalent yields. The farthest sensor was taken as the comparison value but may not have represented the terminal yield value for the highest-energy tests. It was reported that the test results demonstrated that explosive yield varied with contact area. The 300 lb test series (9 each) varied the oxidizer-to-fuel mixture ratios in order to observe the effect on explosive yield. The mixture ratios chosen for $N_2O_4$/A-50 were 1:1, 2:1, and 3:1. $N_2O_4$/A-50 yields varied from 0.4% to 36% TNT as a function of contact area.

#### 4.2.2.2 Simulated Tankage Tests

A separate follow-up test series was conducted at the conclusion of the 300 lb maximum yield tests [43]. The follow-up program was intended to investigate the blast and fireball characteristics of hypergolic versus cryogenic propellants under simulated space booster launch conditions. The test setup involved using scaled propellant tankage configurations with failure modes caused by loss of thrust (two each) or a powered fallback (two each). The propellant tanks were loaded with 300 lbs of total weight for each test to compare test results directly with the previous mixing tests. The tankage L/D chosen was 1.6 for $N_2O_4$/A-50 and 1.8 for $LO_2$/RP-l. Propellant mixture ratios were also fixed at 2:1 for $N_2O_4$/A-50 and 2.5:1 for $LO_2$/RP-1. The static failure mode was accomplished by cutting a common tank bulkhead using a linear-shaped charge. Fuel was in the upper tank, and oxidizer was contained in the lower. In the fallback mode, the tankage was dropped from a 15 ft height and the inter-tank bulkhead ruptured by a ram. For this test, the oxidizer was in the upper tank, fuel in the lower. An explosive charge with time delay was used to initiate detonation. Two time delays were used (0.1 sec and 0.4 sec) to test the effect of time delay on yield. As in the previous tests, data were obtained for peak overpressure, positive impulse, and fireball history with results referenced to TNT. Instrumentation location was also similar to the previous tests. It was reported that typical blast pressure-time histories were obtained for all cryogenic tests; however, the hypergolic yields were so low that no pressure trace was discernable in two of the static tests. $N_2O_4$/A-50 yields varied from << 1% to 1% TNT as a function of contact area.

### 4.2.3 Aerojet-General (Detonation Tests)

In 1964, Aerojet-General conducted experiments to determine whether, under ideal conditions of vigorous mixing, the hypergolic propellant combination of $N_2O_4$ and A-50 could detonate. It was postulated that if a failure mode could be found such that rapid mixing occurred prior to the finite ignition time delay of the hypergolic components, then a mass detonation of the liquid mixture would result [44]. To test this hypothesis, a technique was devised to intimately disperse one of the hypergolic propellants rapidly within the other. Various mixing techniques were studied experimentally to establish an optimum configuration. The mixing method that was determined to give the best results was the concentric vessel technique in which a glass cylinder was filled with one liquid propellant, then placed concentrically within a larger cylindrical vessel containing the other propellant. The inner vessel was shattered by initiating a pair of primacord strips 180° apart. Following a preplanned time delay, a second pair of primacord strips, located 90° from the first, were initiated to provide the detonating stimulus for the mixed propellants. The tests were conducted in an oxidizer vat containing eleven 4 ft plexiglass tubes containing fuel. The tubes were shattered using a mild detonating fuse prior to initiating a planar booster at the base of the vat. Propellant quantities varied from 100 lbs to 140 lbs at a mixture ratio of 2.4:1. Twelve tests were performed with initiation delay time varied from 0.1 msec to infinity. No evidence of propagating detonation was noted with the yields varying from 14% TNT to 52% TNT.

### 4.2.4 Aerojet-General (Apollo)

In 1965, Aerojet-General conducted experiments that supported NASA/Manned Spacecraft Center hazard assessments for the Apollo program [48]. The objective was to reduce the uncertainty in quantifying the explosive potential by controlling/minimizing the propellant mixture. Previous tests were performed by Aerojet-General in 1963 and 1964 for the Gemini program using $LO_2$/RP-1 and $N_2O_4$/A-50, and additional $LO_2$/RP-1 and $N_2O_4$/A-50 tests were performed during this Apollo test series. In these experiments, an external shock was transmitted through the metal pan to rupture the glass dewars, resulting in instantaneous mixing of the propellants. A single $N_2O_4$/A-50 test was performed at 230 lbs that resulted in a 42% to 52% TNT explosive yield.

## 4.3 Rocketdyne

A test program was performed [33][34] to develop safety and design criteria for five storable liquid propellants. The propellants involved in the investigation were $N_2O_4$, $ClF_3$, $N_2H_4$, pentaborane ($B_5H_9$), and Aerozine-50 (50% $N_2H_4$ + 50% UDMH). The hazards program in [33] consisted of both small- and large-scale spill tests of various mono- and bipropellant combinations to investigate both explosive and toxic hazards. The intent of the hazard evaluation effort was to determine the hazards involved in storage and handling of these propellants in quantities up to 5,000,000 lbs. Small-scale tests were intended to simulate failures, such as line leakage or failure, and large-scale tests were intended to simulate tank rupture. Safe-distance storage assessments were also made for the four propellant types. Additional investigations included evaluation of meteorological influence and biological hazards relative to toxic effects. The test project in [34] also performed a series of sub-scale tests of the Titan II missile to determine the hazard potential of the new hypergolic propellants $N_2O_4$/A-50. Missile failures were simulated above ground and in-silo, as well as small-scale and large-scale spill tests of up to 1,600 lbs. The test program was performed in 1960 and 1961.

### 4.3.1 Small-Scale Spill Tests

There were 23 $N_2O_4/N_2H_4$, 13 $B_5H_9/N_2H_4$, 13 $ClF_3/N_2H_4$, 12 $ClF_3/B_5H_9$, 5 $N_2O_4/ClF_3$, and 5 $N_2O_4/B_5H_9$ small-scale tests ranging from 1.7 lbs to 19 lbs. Several tests were conducted without instrumentation. The largest explosions that were recorded during the small-scale test program were caused by combined spills of $B_5H_9/N_2H_4$ on concrete, dirt, and asphalt. The explosions occurred as a single shock wave with reflections and were originated at times varying from slightly after ignition to the start of the post-test purge. Strong overpressures were recorded on five of seven instrumented spills and were observed on four of six un-instrumented tests. $N_2O_4/B_5H_9$ spills were conducted without blast instrumentation; therefore, no TNT equivalents were calculated. However, it was concluded that this combination was potentially the most hazardous of the combinations tested. The $N_2O_4/ClF_3$ combination was not hypergolic as would be expected.

### 4.3.2 Large-Scale Spill Tests

The magnitude of the hazards involved during leakage or rupture of propellant systems containing pentaborane, hydrazine, chlorine trifluoride, and nitrogen tetroxide was determined by performing controlled spills of large quantities of propellants. A total of nine tests were conducted: five singular and three multiple spill tests and one propellant heating test. The propellants were spilled into a 20 ft × 20 ft × 2 ft steel tray from 150- or 165-gallon tanks. The propellants were rapidly expelled from the tanks through 6-inch ports in less than 2 seconds. The oxidizer/fuel tests were conducted using 1,600 lbs of $N_2O_4/N_2H_4$ at a mixture ratio of 1.29:1 and 1,000 lbs of $ClF_3/N_2H_4$ at a mixture ratio of 1:1. A singular spill test of 500 lbs of pentaborane resulted in ignition upon contact with air, resulting in an intense fire lasting several minutes. A combined spill test was also conducted using 275 lbs of pentaborane and 100 lbs of $N_2H_4$. Several large explosions were recorded during this spill test. In the heating test, 135 lbs of hydrazine were heated inside a stainless-steel tank with 90% ullage for 5 minutes, resulting in a hydrazine explosion that fragmented the tank and scattered pieces over a radius of 1,100 feet.

### 4.3.3 Titan II Tests

The Titan II test program involved 9 small-scale (2.5 lbs), 20 sub-scale model (50 lbs and 300 lbs), and 2 large-scale (1,300 lbs and 1,600 lbs) $N_2O_4$/A-50 tests. The small-scale tests used propellant feed lines to inject the liquids into a reaction basin, whereas the model tests used 1/18- and 1/10-scale propellant tanks stacked vertically. The model tests were initiated by removing the bottoms from both tanks either simultaneously or sequentially above ground into a reaction basin or below ground into an open silo. The two large-scale tests used dump tanks and did not simulate a vehicle configuration. The 1,600 lb test was conducted in the presence of deluge water and no overpressure was registered. All data were in terms of TNT equivalence. Only peak overpressure data were recorded. The Titan II model tests varied the spill sequence and surface, including water-covered surfaces in some cases.

## 4.4 Atlantic Research Corporation

In 1961 and 1962, Atlantic Research Corporation investigated the explosive mechanism from hydrazine-type liquids when in contact with liquid nitrogen tetroxide during testing performed for the Air Force Flight Accessories Laboratory at the Aeronautical Systems Division [40]. All testing was done using $N_2O_4/N_2H_4$ at laboratory scale (1 cc to 10 cc) and included falling droplet, liquid interface injection, and gas-liquid experiments. The tests yielded only qualitative data in terms of explosive effects since only photographic instrumentation was used. A series of dump tests were also reported using up to 700 ml

(2 lbs) of reactants. Data from the dump tests were not reported; however, information in the report implies yields on the order of less than 0.1% TNT equivalency. Although of interest from a reaction-mechanism perspective, these tests were not found useful in assessing vehicle-scale explosive risks and are not included in the data assessment. However, findings included explosive correlations with contact-sequence and drop-height dynamic interaction. The investigators also concluded that, contrary to the Rocketdyne test report discussed in section 4.3, the $N_2O_4/N_2H_4$ explosions did not result from vapor-phase hydrazine/air reaction but rather from the presence of the more reactive oxidizer, even though a final conclusion regarding reaction mechanics at the interface could not be defined.

## 4.5 The Martin Marietta Corporation

The Martin Marietta Corporation conducted a series of Titan vehicle destruct tests in support of the Titan II/Dyna-Soar Program [39]. The tests were performed under contract to the Air Force Space Systems Division and used $N_2O_4$/A-50. Small-scale tests used propellant weights up to 292 lbs. Two large-scale tests were also conducted with propellant weights of 32,700 lbs and 15,800 lbs, which simulated half-scale Titan Stages 1 and 2, respectively. The test project took place in 1961 and 1962. Since the test program was intended to develop a destruct system for the Titan vehicle, emphasis was not placed on maximizing and measuring explosive effects. Instead, the tests principally concentrated on destruct charge placement, configuration, and mixing phenomenon. Small-scale test configurations included vertical and horizontal vehicle simulations with destruct charges located along the tank barrels, on the tank hemispherical bulkheads, or at both locations. The half-scale test simulated the Titan II vehicle tank configurations and was oriented vertically. The stage 1 length/diameter (L/D) was 6, while the stage 2 L/D was 3.4. The half-scale test destruct charges were placed on both hemispherical tank domes in the intertank area and represent a confined-by-missile (CBM) type of mixing mode. The small-scale tests yielded only qualitative data in terms of explosive effects because no instrumentation was used. Post-test observations were visual. The half-scale tests were instrumented with pressure and temperature sensors. Due to a test malfunction, no data were obtained for the stage 2 test; therefore, the only usable data from this test series applicable to propellant explosive hazards were obtained from the stage 1 test. The stage 1 destruct test resulted in two detonations with yields that did not exceed 2% TNT equivalency.

## 4.6 URS Corporation (Project PYRO)

Project PYRO was the largest of the liquid-propellant explosion studies. This was an extensive experimental project conducted in the early- and mid-1960s to establish the blast environment for the three common liquid propellant combinations ($LO_2$/RP-1, $LO_2/LH_2$, and $N_2O_4$/A-50). The PYRO $N_2O_4$/A-50 experiments were conducted in two phases [46][47][50]. Propellant quantities ranged from small (200 lbs) to medium-scale (1,000 lbs) at 1.9:1 mixture ratio. In total, 32 successful $N_2O_4$/A-50 tests were performed. The test series for the hypergolic tests were subdivided by configuration and failure mode. The configurations were defined as static test stand, launch pad (pre-launch and launch), and post-launch (in-flight and ground impact). The series of postulated failure modes were identified as CBM (3 tests), confined-by-ground surface (CBGS) (2 tests), tower drop (11 tests), explosive donor (2 tests), command destruct (2 tests), and high-velocity impact (HVI) (10 tests). Three additional tests were attempted using chilled propellants. The PYRO CBM apparatus was similar to the Marshall Space Flight Center (MSFC) CBM apparatus, consisting of a single tank separated by a glass diaphragm. However, in the PYRO tests, the diaphragm was broken using a cutter ram driven by an explosive charge. In the CBGS experiments, the glass diaphragm in the tank was broken by dropping the tank from a tower onto a cutter ram. An additional special cold-propellant test case was added in [49] to determine whether a mixing technique could be used to increase the explosive yield of hypergolic propellants.

### 4.6.1 Static Test Stand Case

Failure mode tests included CBM, CBGS, small explosive donor, and 100 ft tower drop. The range of estimated upper limit yield was determined to be from 0.5% TNT (CBGS) to 1.5% TNT (CBM) to 2% TNT (small donor).

### 4.6.2 Launch Pad Case

Failure mode tests included CBM, CBGS, small and large explosive donor, command destruct, and 300 ft (~140 ft/sec) fallback. The range of estimated upper-limit yield was determined to be from 0.5% TNT (CBGS and command destruct) to 3% TNT (fallback) to 5% TNT (large donor). As reported in Project PYRO, for the launch pad case, the highest yields were obtained from the large explosive-donor case. However, an explosive donor weighing two or three times the effective weight of the exploding propellant mass was required to achieve this large a yield. Thus, in any case where this situation occurs, there would be more concern about the blast from the donor than that from the resulting propellant explosion. Therefore, it seemed reasonable to present two yield values for this case: one with a large explosive donor and one without.

### 4.6.3 Post-Launch Case

Failures after launch are divided into two subcategories: in-flight and HVI ground impact. In-flight failure modes included CBM, small and large explosive donor, and command destruct. Ground impact modes included flat-surface and soft-surface impact cases. The worst credible failure mode for the post-launch case is the high-velocity ground impact. This failure mode would occur from high-altitude fallback or powered impact. The data from the experimental tests indicate that at the maximum impact velocity investigated (~570 ft/sec), the mean of the explosive yields varied from 14% (for impact on a flat surface) to ≥ 80%, depending on the degree of cratering and therefore containment provided by the target surface. The estimated upper limit for these two failure modes is 24% for the flat-surface case and 96% for the cratering case.

### 4.6.4 Cold-Propellant Case

The oxidizer and fuel were cooled in separate containers, using $LN_2$ to chill the propellants before mixing. Three tests were attempted using 200 lbs of $N_2O_4$/A-50 with yields of 6.1% TNT to 13% TNT from 2 of the tests.

## 4.7 NASA White Sands Test Facility (WSTF)

Beginning in the early 1990s, a series of propellant tests was performed at the NASA WSTF. The tests were separated into three programs. The first was the Large-Scale Hydrogen/Oxygen Explosion (LSHOE) test program, the second was the Propellant Reaction Characterization (PRC) test program, and the third was the Hydrogen/Oxygen Vertical Impact (HOVI) test program. PRC testing was an extension of the $LO_2$/$LH_2$ distributed mixing tests performed by LSHOE to characterize the maximum yield from other possible propellant mixtures resulting from a Titan IVA/Centaur failure. The test programs included development of hydrodynamic mixing and reaction models. LSHOE conducted a series of distributed mixing tests of medium scale (150 lbs and 500 lbs). The PRC tests included immersion testing of 150 lb mixtures of $LO_2$/MMH, $N_2O_4$/MMH, $N_2O_4$/$LH_2$, $LO_2$/RP-1, and $H_2O_2$/Jet-A. Three $N_2O_4$/MMH tests were performed at 150 lbs [54] and a weight ratio of 2.8:1. The $N_2O_4$/MMH PRC tests yielded TNT equivalences of 34% to 66%.

# 5. Hypergolic Explosive Assessment

## 5.1 Assessment Methodology

Due to observed differences in explosive yield assessment methods used by the various experimenters and analysts, as part of the original study [6], all liquid-propellant explosion data were analyzed using a consistent methodology. The approach used was as follows:

- Data values were corrected as needed where calibration data were available
- Identified instrumentation faults were discarded
- Yields were calculated in terms of overpressure
- Far-field measurements were averaged
- All data were recalculated using Kingery's surface-burst factors to establish TNT percent yields

The TNT percent yields derived for all tests and failures were then compiled by mass and mixing scenario. To accomplish this, the following mixing scenarios were defined:

STATIC MIXING

- Propellant spills from tank rupture/leakage
- Inter-tank bulkhead rupture
- Vehicle destruct test cases included
- Static conditions ≤ 10 ft/sec

DYNAMIC MIXING

- Vertical fallback impact
- Horizontal fallback impact (toppling)
- Vehicle destruct flight failures not included
- Dynamic conditions ≤ 150 ft/sec

HIGH-VELOCITY IMPACT MIXING

- Powered impact
- HVI conditions > 150 ft/sec

DISTRIBUTED (IMMERSION) MIXING

- Matrix of propellant immersed within a bath

DONOR MIXING

- Secondary reaction resulting from initial (smaller) explosion
- Data included in dynamic mixing

## 5.2 Hypergolic Vehicle Failures

Results of the independently performed explosive yield analyses from data obtained from [58]–[73] for the hypergolic liquid-propellant launch vehicle failures are tabulated in Table 3 by vehicle. Included are the propellant type and quantity, failure mode, and estimated explosive yield resulting from the failure reaction. The Long March failure was analyzed as total vehicle yield based on the reconstruction of the accident scenario ([6], Addendum 1).

Table 3. Hypergolic Launch Vehicle Failures

| VEHICLE | MODE | PROPELLANTS | K lb | YIELD |
|---|---|---|---|---|
| LONG MARCH CZ-3B[1] | Impact | $N_2O_4$/UDMH<br>$LO_2$/$LH_2$ | 776<br>39 | 12.5% |
| TITAN II | | | | |
| Titan II B-54 | Fallback | $N_2O_4$/A-50 | 230 | 1.0% |
| Titan II B-22 | Fallback | $N_2O_4$/A-50 | 230 | 2.0% |
| Titan II B-57 | Donor | $N_2O_4$/A-50 | 230 | UNK |
| Titan II M68B-25 | Donor | $N_2O_4$/A-50 | 230 | 3.0% |
| TITAN III | | | | |
| Titan 34D-9 | Destruct | $N_2O_4$/A-50 | 384 | 1.5% |

[1]Combined yield

## 5.3 Hypergolic Propellant Data Assessment

The plots in Figure 7 through Figure 16 are the results of this assessment. Figure 7 and Figure 8 include all hypergolic propellant data points ($N_2O_4$/A-50, $N_2O_4$/$N_2H_4$, $N_2O_4$/MMH, $N_2O_4$/UDMH, PB/$N_2H_4$, CTF/$N_2H_4$, and CTF/PB) whether from tests or failures. Each dataset is identified by mixing mode. The data were then assessed for large-scale (≥ 100 lb) conditions and the 95% upper limits established for each dataset (i.e., static or dynamic mixing) for each propellant combination.

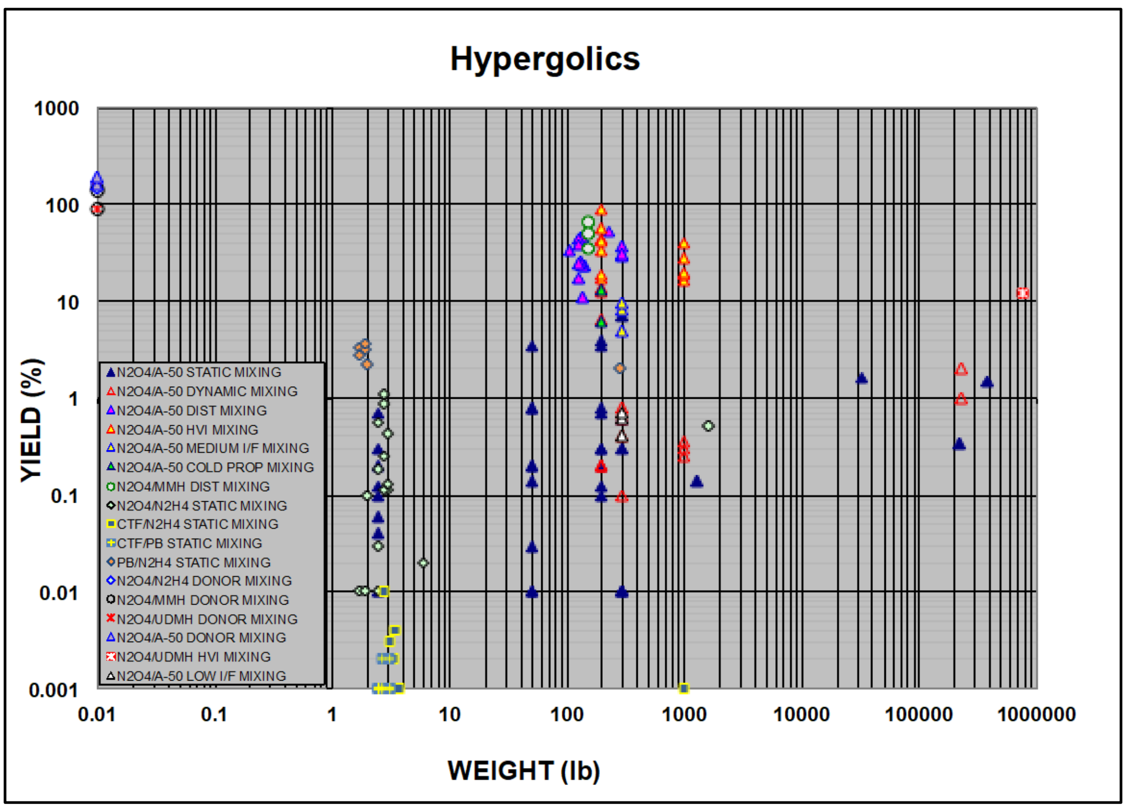


Figure 7. EEW yields for hypergolic propellants.

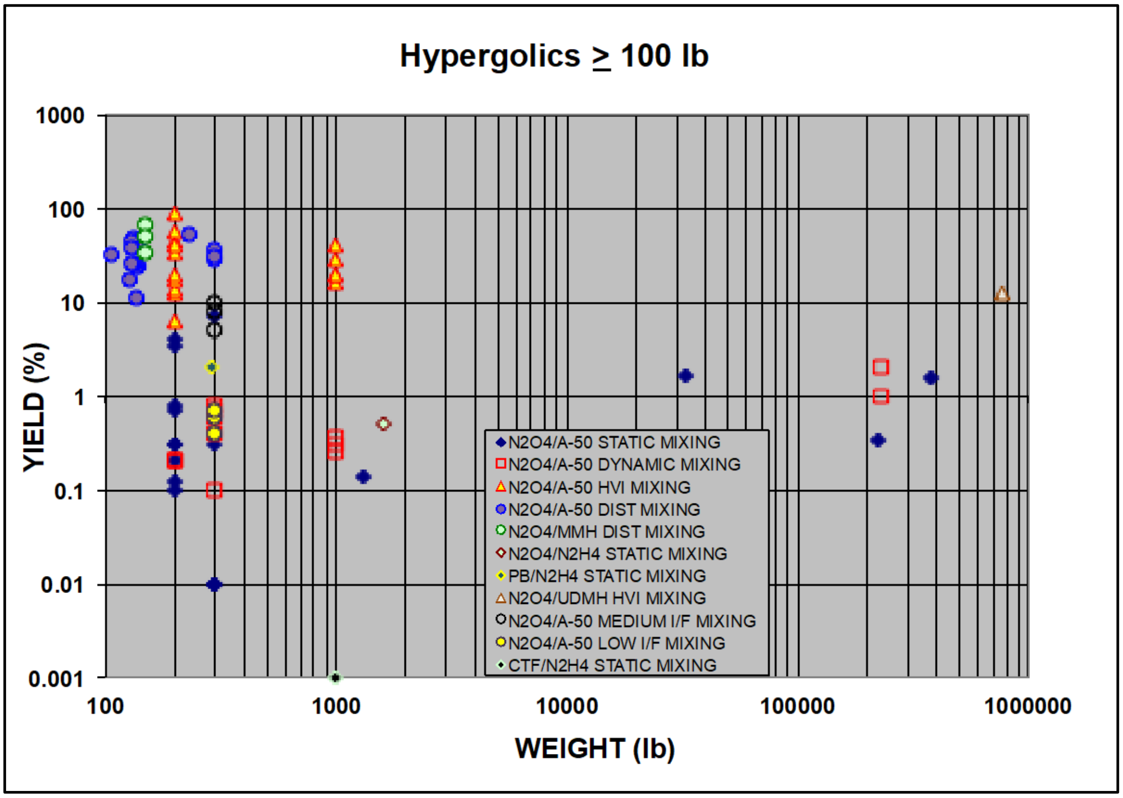


Figure 8. EEW yields for hypergolic propellants ≥ 100 lbs.

### 5.3.1 $N_2O_4$/A-50 Data

Figure 9 shows all $N_2O_4$/A-50 data, and Figure 10 includes the 95% upper-limit level for static and dynamic mixing modes for the data ≥ 100 lbs.

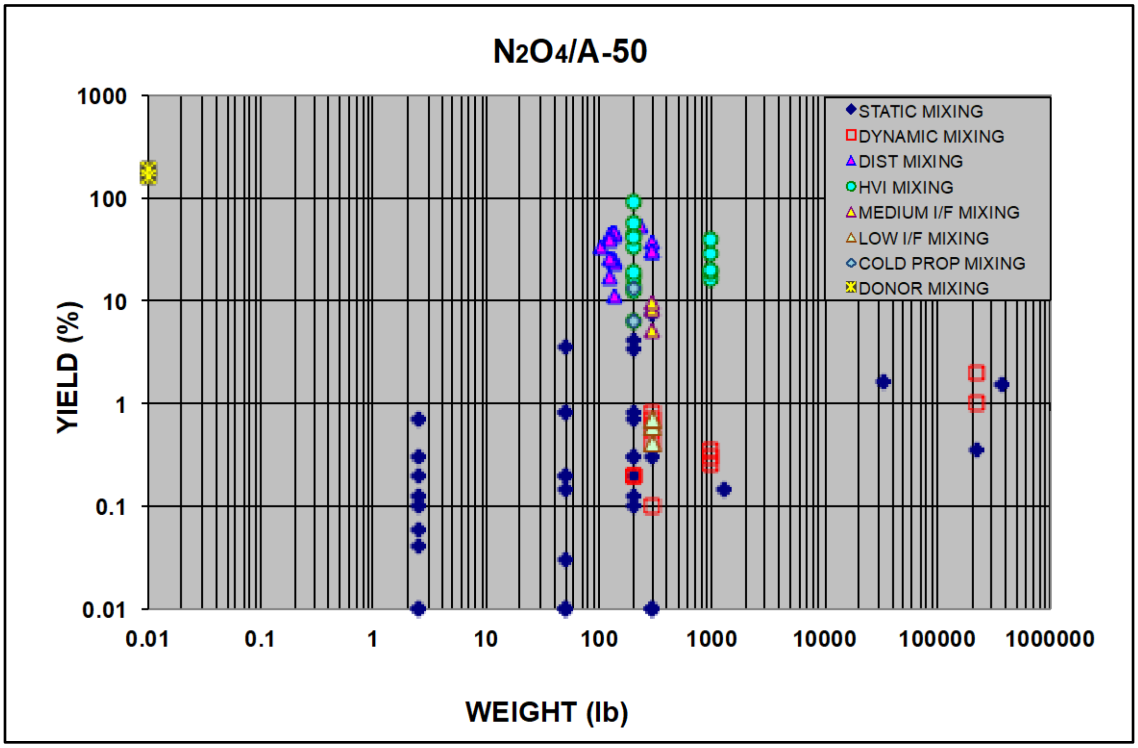


Figure 9. EEW yields for all $N_2O_4$/A-50 tests/failures.

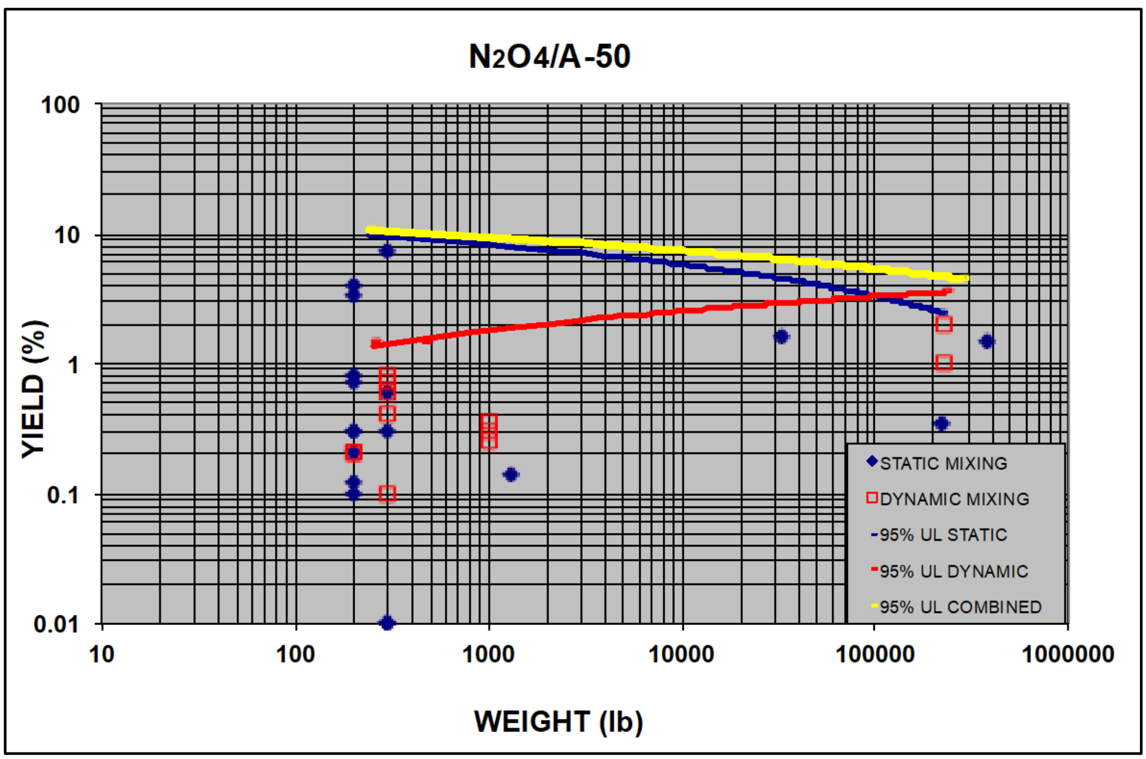


Figure 10. 95% upper limit for $N_2O_4$/A-50 EEW yields.

### 5.3.2 $N_2O_4/N_2H_4$ Data

Figure 11 shows all $N_2O_4/N_2H_4$ data obtained to date along with the $N_2O_4$/A-50 data. Figure 12 compares the ≤ 5 lb $N_2O_4/N_2H_4$ data 95% upper-limit values to similar $N_2O_4$/A-50 data in conjunction with the $N_2O_4$/A-50 ≥ 100 lb data and 95% upper-limit plots.

### 5.3.3 $N_2O_4$/MMH and $N_2O_4$/UDMH Data

Figure 13 shows all $N_2O_4$/MMH and UDMH data obtained to date along with the $N_2O_4$/A-50 data. Figure 14 compares the ≥ 100 lb $N_2O_4$/MMH DMT data 95% upper-limit values to similar $N_2O_4$/A-50 data in conjunction with the $N_2O_4$/A-50 ≥ 100 lb data and 95% upper-limit plots.

### 5.3.4 PB and CTF Data

Figure 15 shows all PB/$N_2H_4$, CTF/$N_2H_4$, and CTF/PB data obtained to date along with the $N_2O_4$/A-50 data. Figure 16 compares the ≤ 5 lb PB and CTF data 95% upper-limit values to similar $N_2O_4$/A-50 data in conjunction with the $N_2O_4$/A-50 ≥ 100 lb data and 95% upper-limit plots.

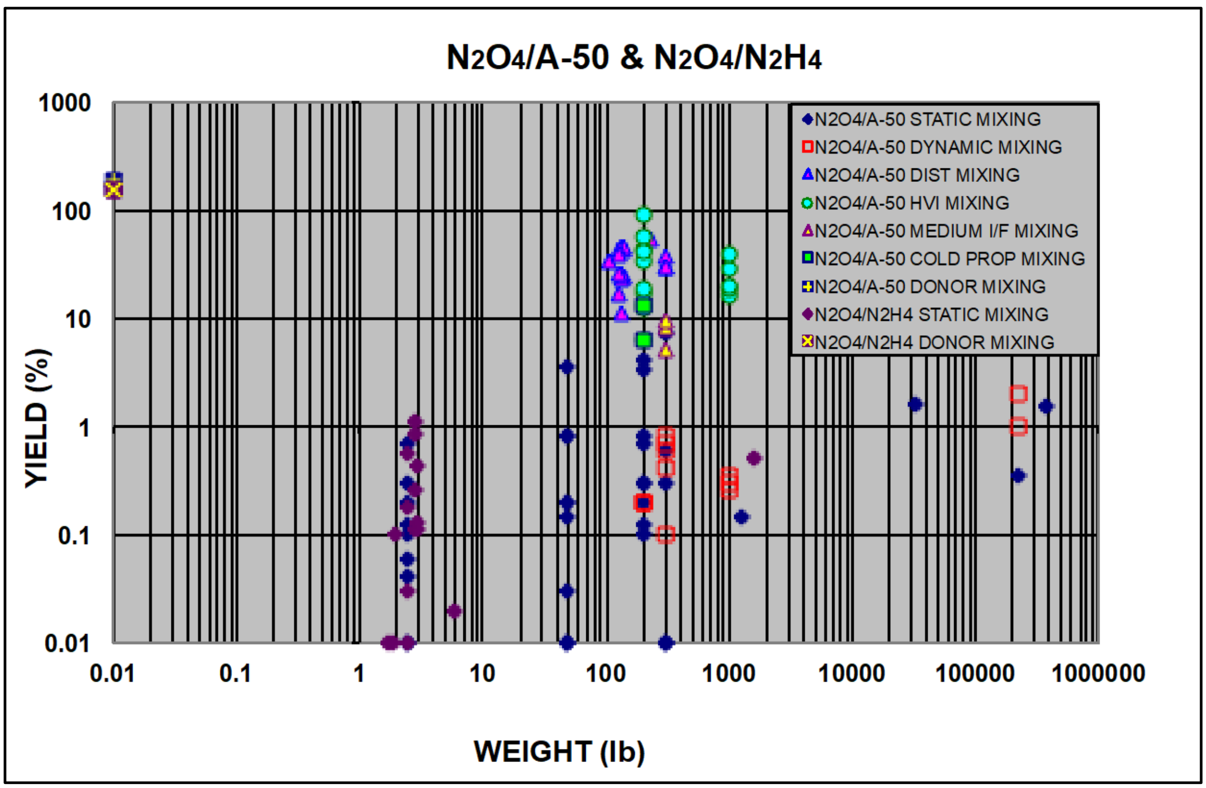


Figure 11. EEW yields for all $N_2O_4/N_2H_4$ tests.

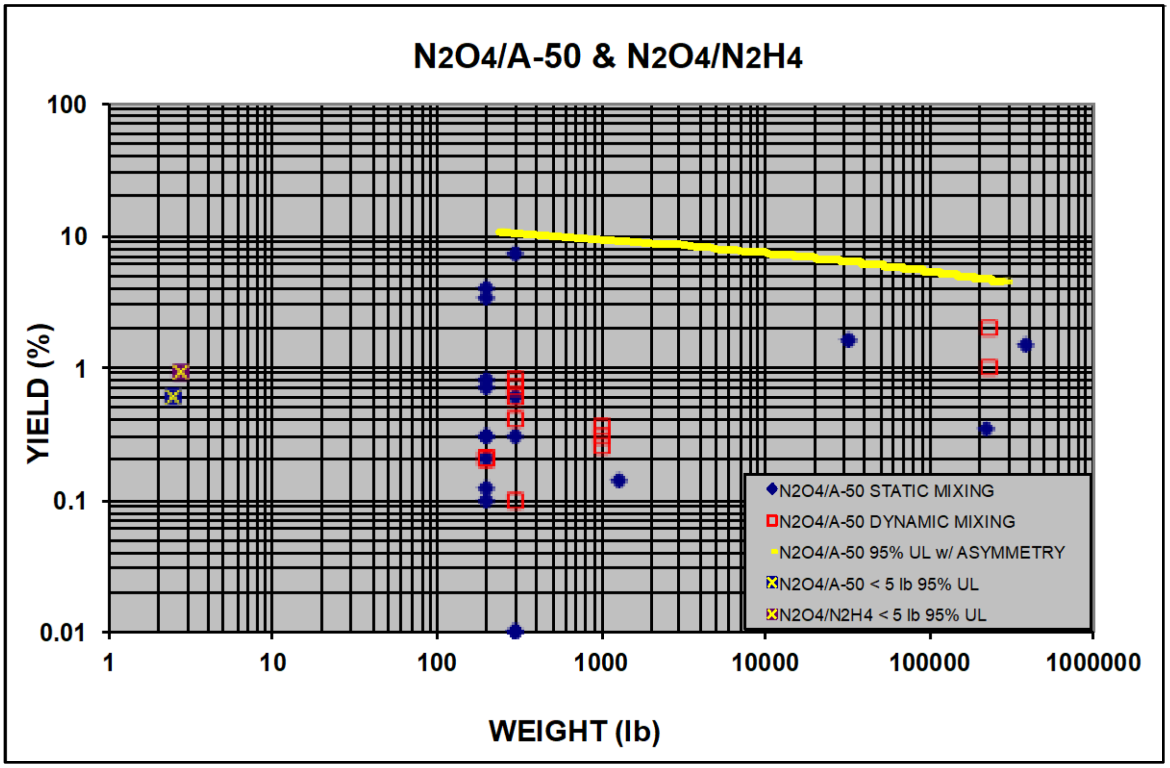


Figure 12. 95% upper limit for $N_2O_4$/A-50 and comparable $N_2O_4/N_2H_4$ EEW yields.

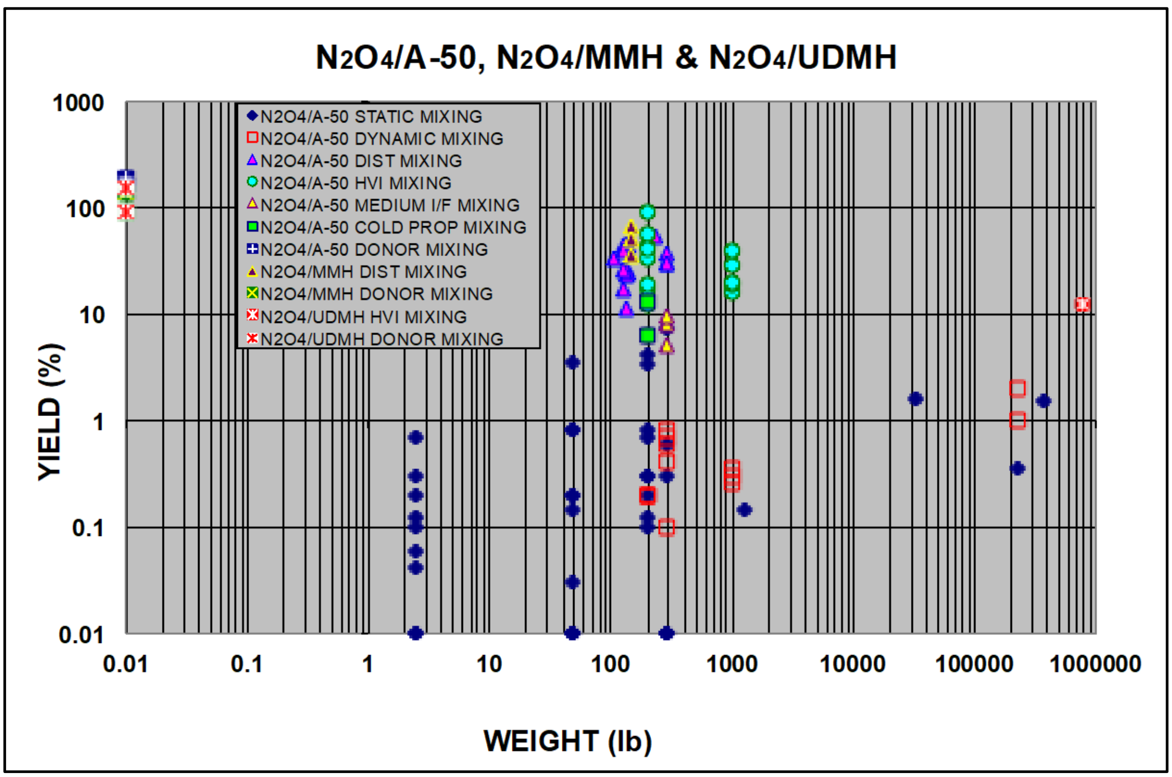


Figure 13. EEW yields for all $N_2O_4$/MMH and $N_2O_4$/UDMH tests/failures.

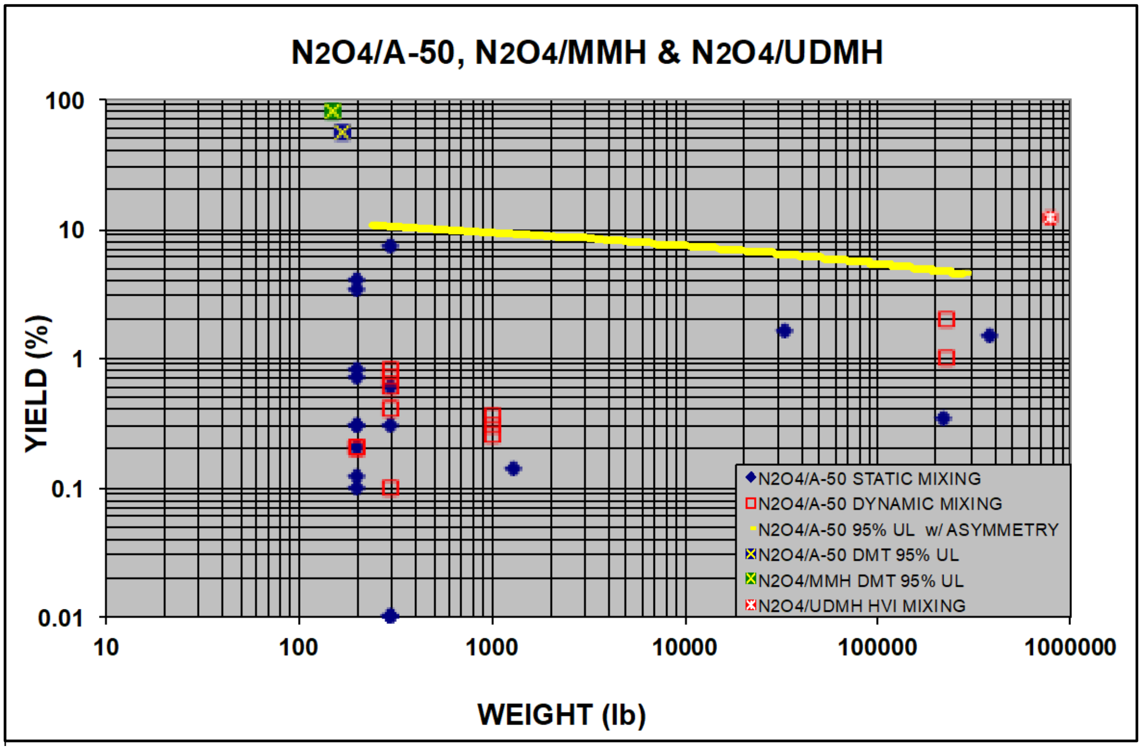


Figure 14. 95% upper limit for $N_2O_4$/A-50 and comparable $N_2O_4$/MMH EEW yields.

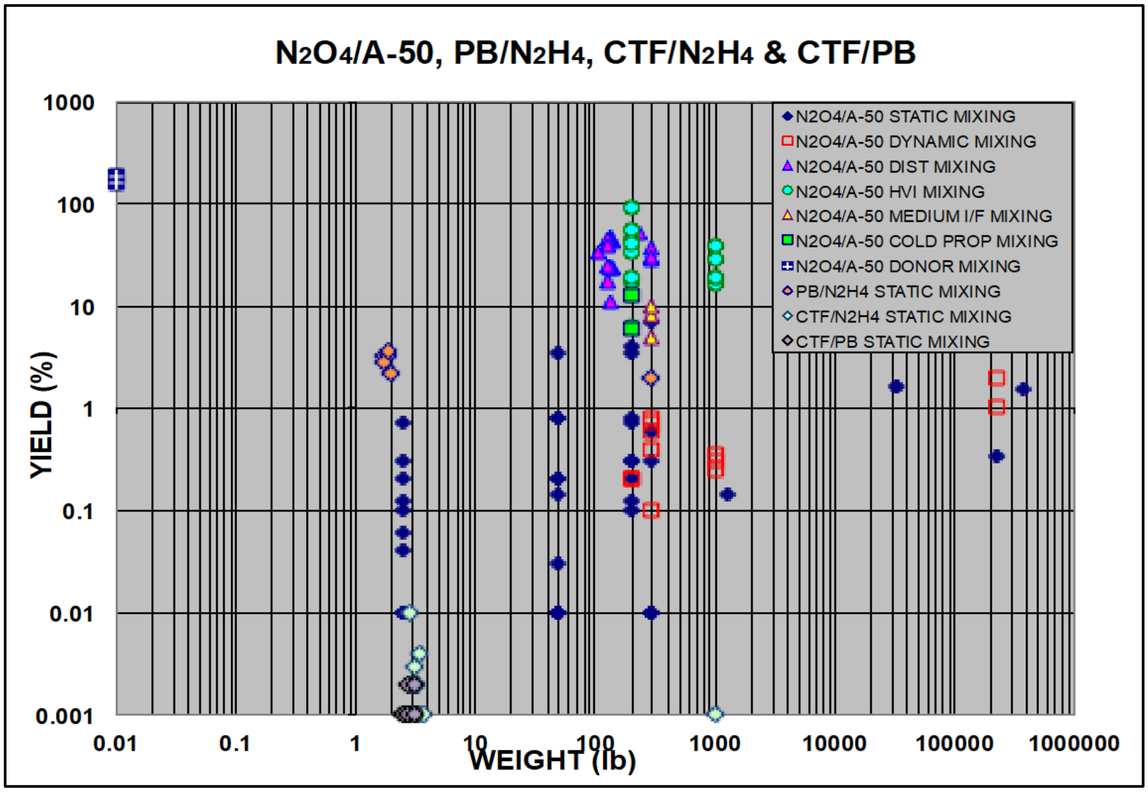


Figure 15. EEW yields for all PB and CTF tests.

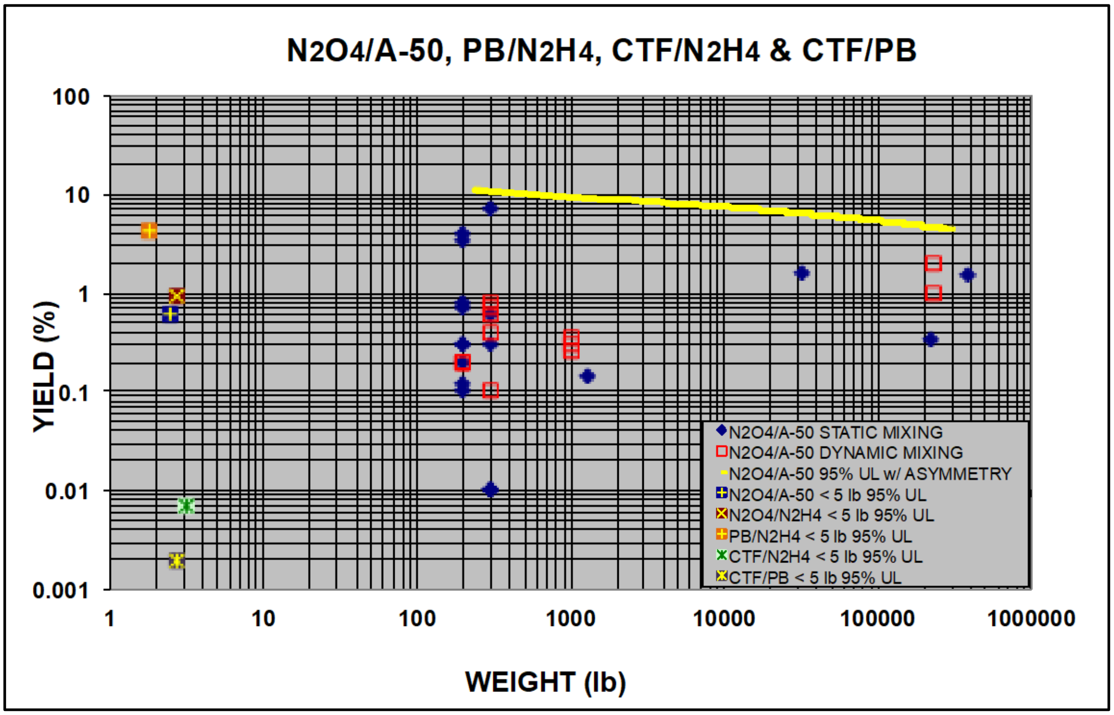


Figure 16. 95% upper limits for PB and CTF and comparable $N_2O_4/N_2H_4$ EEW yields.

### 5.4 Discussion

As mentioned above, since the initial assessments presented in [3], several adjustments have been included in this analysis. One modification was to include additional propellant combinations rather than addressing the $N_2O_4$/A-50 propellant combination as a single assessment. Most of this data were initially included in [6] but not assessed in [3]. Additional test reports and failure data were obtained and incorporated as well. The plots shown in previous sections reflect this revised assessment.

#### 5.4.1 $N_2O_4$/A-50 Results

In this report, $N_2O_4$/A-50 Yield Curves were created using the dataset shown in Figure 9. Each yield curve was created by converting the percent yield versus propellant weight plots shown in Figure 10 into TNT weight ($W_T$) versus propellant weight ($W_P$) plots for both the static and dynamic 95% upper-limit cases and performing curve fits.

The resulting yield curves, shown in Figure 17, are $W_T = 0.20\ W_P^{7/8}$ for static and $W_T = 0.25\ W_P^{7/8}$ for dynamic conditions. The yield curves are shown along with the current DESR 6055.09 Table V5.E4.T5 [1] values. The results are conclusive in that the vehicle-scale quantity dynamic yields are significantly less than the current values.

A-50 was developed for use in the Titan II missile and was widely used in all Titan III, Titan IV, and Delta launch vehicles, the Apollo program, and other spacecraft. A-50's 50/50 mixture of $N_2H_4$ and UDMH added the extra performance of hydrazine with the stable characteristics of UDMH.

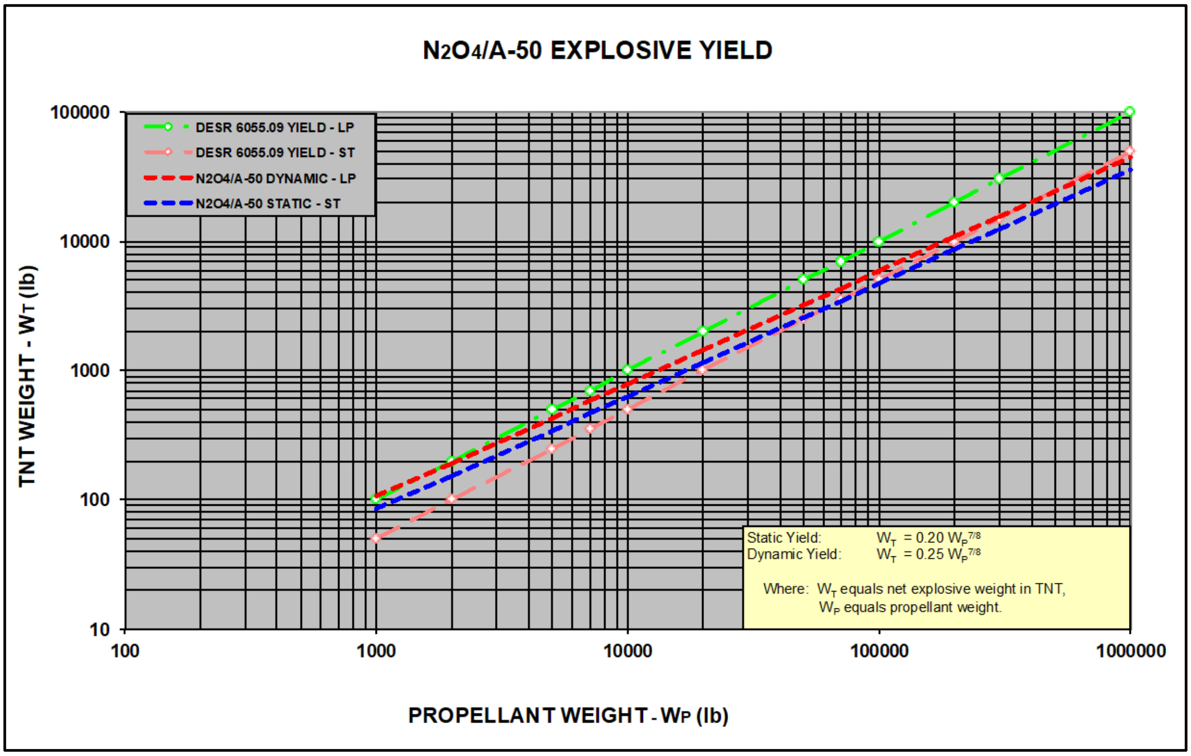


Figure 17. Proposed $N_2O_4$/A-50 Yield Curves and current DESR 6055.09 [1] values

The explosive nature of hypergolic propellant combinations is fundamentally different from any other combination type. Since the explosive yield is directly proportional to the initial contact area, the yields do not correspond to available or mixed volumes or mass. Therefore, the term "mixing" is not relevant to these liquids. This phenomenon can be readily observed in Figure 10 by noting the slight decrease in yield under static conditions as available weight increases while the initial contact area remains relatively fixed. Whereas a moderate yield increase is noted under dynamic conditions as the prompt interface area is instantaneously increased prior to initial reaction. These propellants also exhibit multiple pressure pulses (as opposed to a single reaction) as separate ignition events occur at various locations throughout the "mixing" domain.

While the static yields remain on the order of the current 5%, the dynamic yields are nearly half (~6%) the current 10% at the same vehicle scale. Although large-scale failure data are limited, a reasonable difference in yield factor was found between the static and dynamic cases. This factor was then applied to the overall results to determine static and dynamic ranges. In addition, significant asymmetry was noted in all hypergolic explosive data and an appropriate factor was applied as described in [2], [3], and Appendix A. This asymmetry can be attributed to the distributed nature of the multiple reactions noted in the data. At lower mass-scale conditions, however, both the static and dynamic yields are on the order of the current 10% used only for dynamic launch pad conditions. Therefore, it is believed that the $N_2O_4$/A-50 Yield Curve equations proposed here realistically represent the mass/yield function and are appropriate for use as a new standard.

Another observation regarding $N_2O_4$/A-50 data—and hypergolic combinations in general—is that the yield associated with the DMTs does not reflect consistency with the overall yield equation as they do for the cryogenic ($LO_2$/$LH_2$) and hydrocarbon ($LO_2$/RP-1) combinations (see [2] and [3]). This disparity can be seen in Figure 18, which shows the dynamic yield curves and DMT results for $LO_2$/$LH_2$ and $LO_2$/RP-1 along with the data and $N_2O_4$/A-50 Yield Curves. The small-scale (< 5 lb) data points are also included.

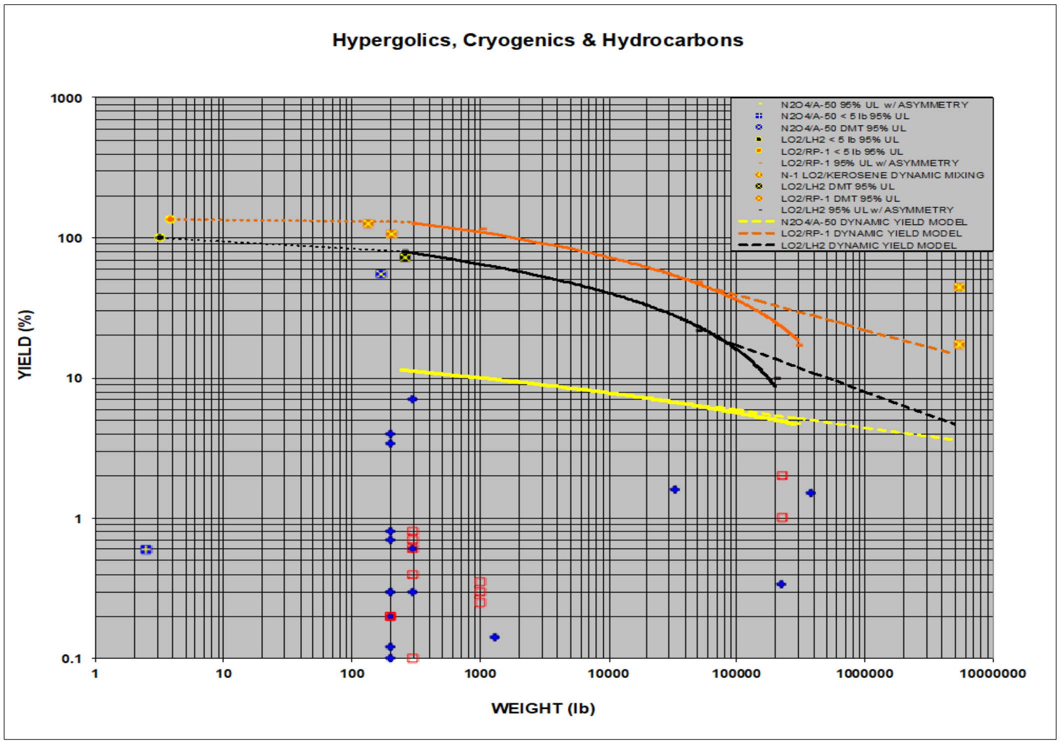


Figure 18. Yield curves, DMT, and < 5 lb data for $LO_2$/RP-1, $LO_2/LH_2$, and $N_2O_4$/A-50.

Figure 19 highlights the differences observed in the cryogenic/hydrocarbon versus the hypergolic explosive data. The DMT observation can be explained by the same reaction phenomenon discussed above regarding the disparity in static and dynamic trends. In the DMT tests, the immediate surface contact areas were varied from low- to high-contact area/lb of propellant. The high-contact area data are plotted here. Variance with yield can be seen in all combinations; however, the increased yield effect due to surface area is most pronounced in the hypergolic reaction. Both the $LO_2$/RP-1 and $LO_2/LH_2$ DMT results correspond to the lower weight range of their respective yield curves. This allows a smooth transition from the static test and launch facility yield functions to a fixed maximum value for quantities below 1,000 lbs at other types of facilities (designated as "other PES" in the criteria).

However, in the $N_2O_4$/A-50 case, this does not occur. A five-fold increase is seen between the lower weights of the yield function and the DMT data. For example, while the yield curve indicates the maximum explosive factor below 1,000 lbs should be on the order of 15%, the DMT data suggest 55% is appropriate. Since it is virtually impossible to create the DMT conditions in any accidental explosion scenario due to the reactive nature of the participants, the DMT results are useful for understanding the hypergolic reaction mechanisms but should not be applied as criteria for any combination. Therefore, in terms of explosive potential, it is proposed that a maximum of 14% TNT should be used for $N_2O_4$/A-50 combinations below 1,000 lbs.

Another contrary result seen in the $N_2O_4$/A-50 comparison is the increased yield with mass that is evident with hypergolics under static mixing conditions. This again is a function of the surface area/lb of propellant increasing significantly under static conditions up to an optimum that may occur on the order of 500 lbs total weight at typical mixture ratios. This is because even at small scale, most hypergolic combinations exhibit random multiple reaction events, whereas for the other propellant combinations (hydrocarbon and cryogenic), most cases result in single explosive reactions as propellant mixing is observed in all other propellant combinations of these types at this scale.

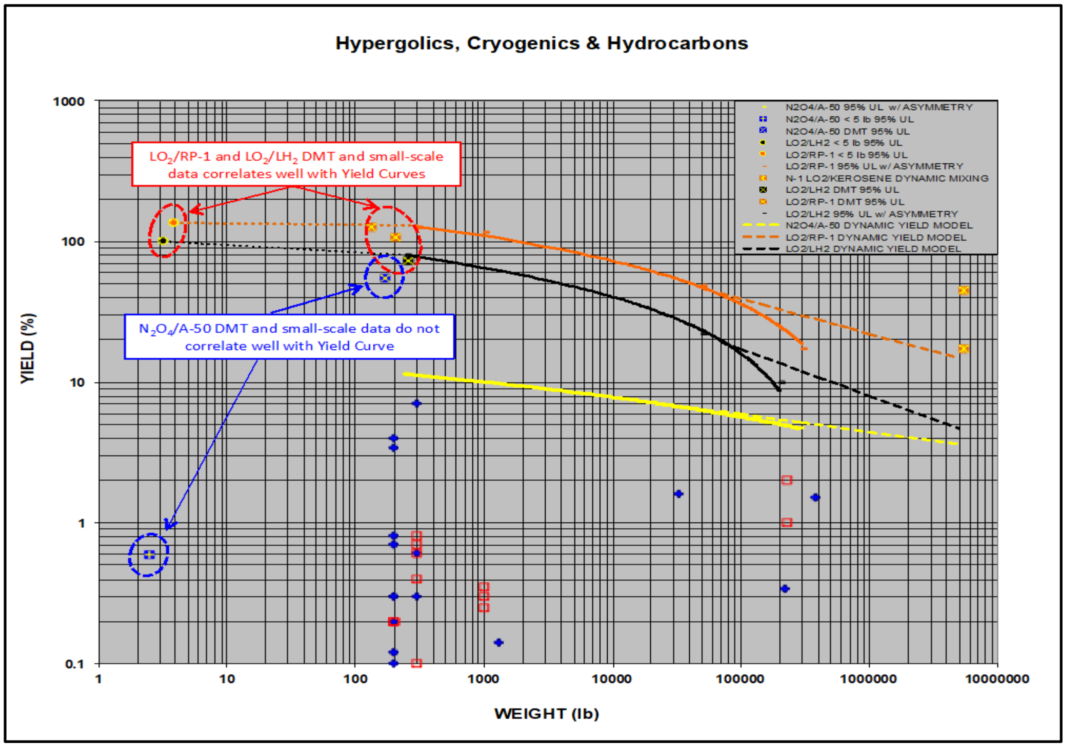


Figure 19. Yield curves, DMT, and < 5 lb data for $LO_2$/RP-1, $LO_2$/$LH_2$, and $N_2O_4$/A-50 with DMT annotation.

An additional observation can be made from the data shown in Figure 20, where the HVI test results for each combination were added to Figure 18. For consistency, the HVI data included in Figure 20 for all cases was from the confined impact case. It can be clearly seen that the $N_2O_4$/A-50 hypergolic combination under these HVI conditions shows a dramatic (~175%) increase over the DMT results. This is attributable to the prompt increase in surface area prior to hypergolic reaction. This increase is similar to that of the cryogenic $LO_2$/$LH_2$ combination, which increases ~208% over the DMT case under HVI dispersion due to the prompt release of the gaseous mixture prior to vapor phase autoignition. In the case of $LO_2$/RP-1, however, the HVI yields actually decrease on the order of 36% when compared to DMT results as mixing is in fact reduced. In this case, the reactants are dispersed under HVI conditions, resulting in a lesser contribution from the available propellants. These impact sensitivity results are discussed in Appendix A and in [2] and [3].

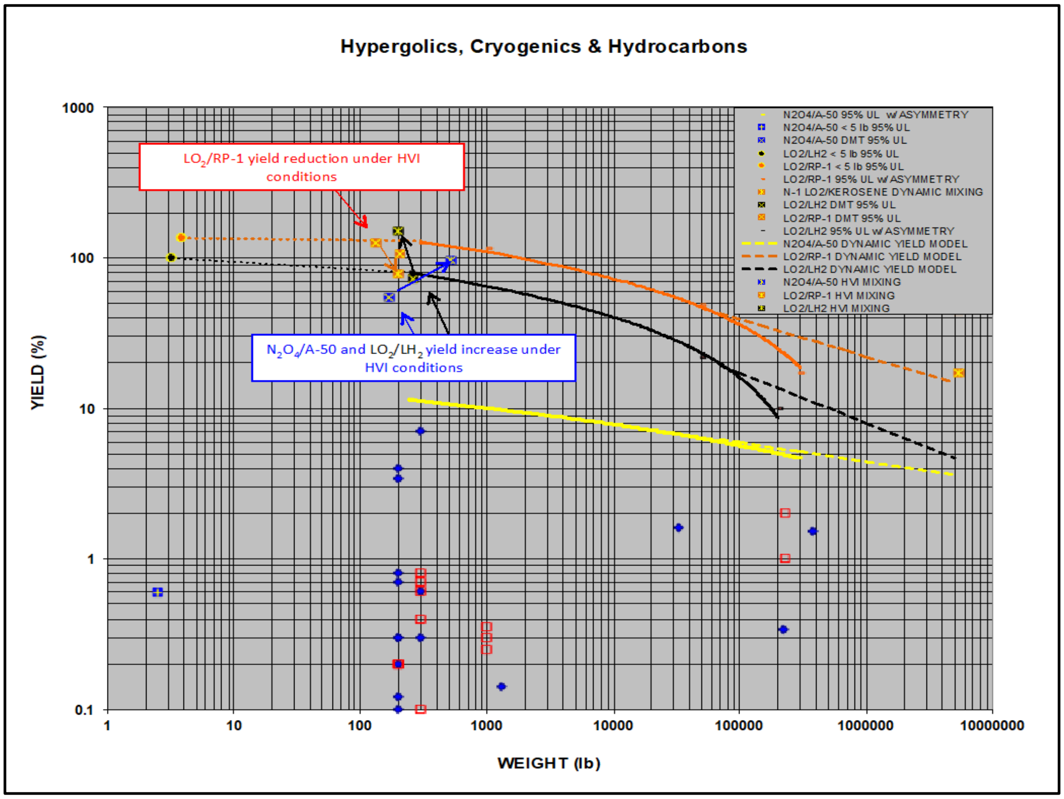


Figure 20. Yield curves, DMT, < 5 lb, and HVI data for $LO_2$/RP-1, $LO_2/LH_2$, and $N_2O_4$/A-50 with HVI annotation.

During Project PYRO, an attempt was made to induce $N_2O_4$/A-50 mixing by first chilling the propellants using $LN_2$. This test data are included in Figure 9. Although the test conditions limit the ability to extract much more than comparative conclusions from the results, it is interesting to note that the yields from these tests were on the order of the DMT medium contact area/lb of propellant results.

### 5.4.2 $N_2O_4/N_2H_4$ Results

$N_2H_4$ saw early use as a hypergolic fuel but was later replaced largely by UDMH. Due to limited test data, a common scale assessment was performed for the $N_2O_4/N_2H_4$ hypergolic combination. Available data are also insufficient to determine any variation between static and dynamic conditions. Hydrazine has been well established as having explosive characteristics in the vapor state [8]. Therefore, the presence of $N_2H_4$ in a bipropellant combination (including the above $N_2O_4$/A-50 combination) enhances the explosive characteristics.

The available data indicate that the $N_2O_4/N_2H_4$ reaction is estimated to be on the order of 25% to 50% more energetic than $N_2O_4$/A-50 due to the additional hydrazine available in the reaction. Using similarity and the data in Figure 11 and Figure 12, a yield curve was created for $N_2O_4/N_2H_4$ that resulted in $W_T = 0.30W_P^{7/8}$ (see Figure 21). It is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 18% TNT should be used for this combination below 1,000 lbs.

$N_2H_4$ remains widely used as a monopropellant but is also used in select spacecraft as an $N_2O_4/N_2H_4$ bipropellant.

### 5.4.3 $N_2O_4$/MMH Results

MMH was developed in the early 1950s at the same time as UDMH. Although MMH had higher performance than UDMH, MMH became less widely used due to production and stability questions. Due to limited test data, a common scale assessment was performed for the $N_2O_4$/MMH hypergolic combination. Although $N_2O_4$/MMH does not have small-scale comparative data, it was also subjected to DMT testing of similar scale to $N_2O_4$/A-50. This combination is observed to be of lower yield than $N_2O_4/N_2H_4$ but slightly greater than $N_2O_4$/A-50. It is acknowledged that this conclusion is limited by data availability and may be somewhat conservative but not significantly so. Using similarity and the data in Figure 13 and Figure 14, a yield curve was created for $N_2O_4$/MMH that resulted in $W_T = 0.28\ W_P^{7/8}$ (see Figure 21). Similar to $N_2O_4/N_2H_4$, it is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 16% TNT should be used for these combinations below 1,000 lbs.

MMH was used by NASA on the Gemini program and was also used on Apollo and on the Space Shuttle. It is currently used on many spacecraft and has largely replaced UDMH due to its higher performance.

### 5.4.4 $N_2O_4$/UDMH and IRFNA/UDMH Results

UDMH was developed in the early 1950s as one of the hydrazine derivatives (MMH, symmetrical dimethyl hydrazine (SDMH), and UDMH). SDMH was discarded, but by 1955, UDMH became the hypergolic fuel of choice throughout the space industry. It was replaced by A-50 or MMH in the U.S. Nitric acid in the form of RFNA, stable fuming nitric acid (SFNA), IRFNA, WFNA, inhibited white fuming nitric acid (IWFNA), etc. was the hypergolic oxidizer of choice in the early missile and launch vehicle programs of the 1940s and 1950s. Nitric acid was replaced by $N_2O_4$ in the 1960s. Due to limited test data, a common scale assessment was performed for the $N_2O_4$/UDMH hypergolic combination; however, no explosive data could be found for IRFNA/UDMH and there is one vehicle failure using $N_2O_4$/UDMH. Although both combinations are observed to be of lower yield than $N_2O_4/N_2H_4$ and $N_2O_4$/A-50, it is acknowledged that this conclusion is limited by data availability. Using similarity and the data in Figure 13 and Figure 14, a yield curve was created for $N_2O_4$/UDMH that resulted in $W_T = 0.19\ W_P^{7/8}$ (see Figure 21). Similar to $N_2O_4$/MMH, it is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 12% TNT should be used for these combinations below 1,000 lbs.

In the case of IRFNA/UDMH, no test or failure data could be found, although the current yield criteria specifies 10% TNT equivalence for both static (test facility) and dynamic (launch facility) conditions in [1], Table V5.E4.T5. However, from a purely reaction chemistry consideration, a difference can be calculated between the yields of $N_2O_4$/UDMH and IRFNA/UDMH. Using relative specific energies, a yield curve was created for IRFNA/UDMH that resulted in $W_T = 0.26\ W_P^{7/8}$ (see Figure 21). Similar to $N_2O_4$/UDMH, it is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 15% TNT should be used for these combinations below 1,000 lbs.

IRFNA/UDMH found continued in use on the Delta and Agena upper stages into the 1970s but is no longer in operational use. There is currently no operational use of $N_2O_4$/UDMH in the U.S., but future use is possible. It remains in use in many non-U.S. hypergolic systems.

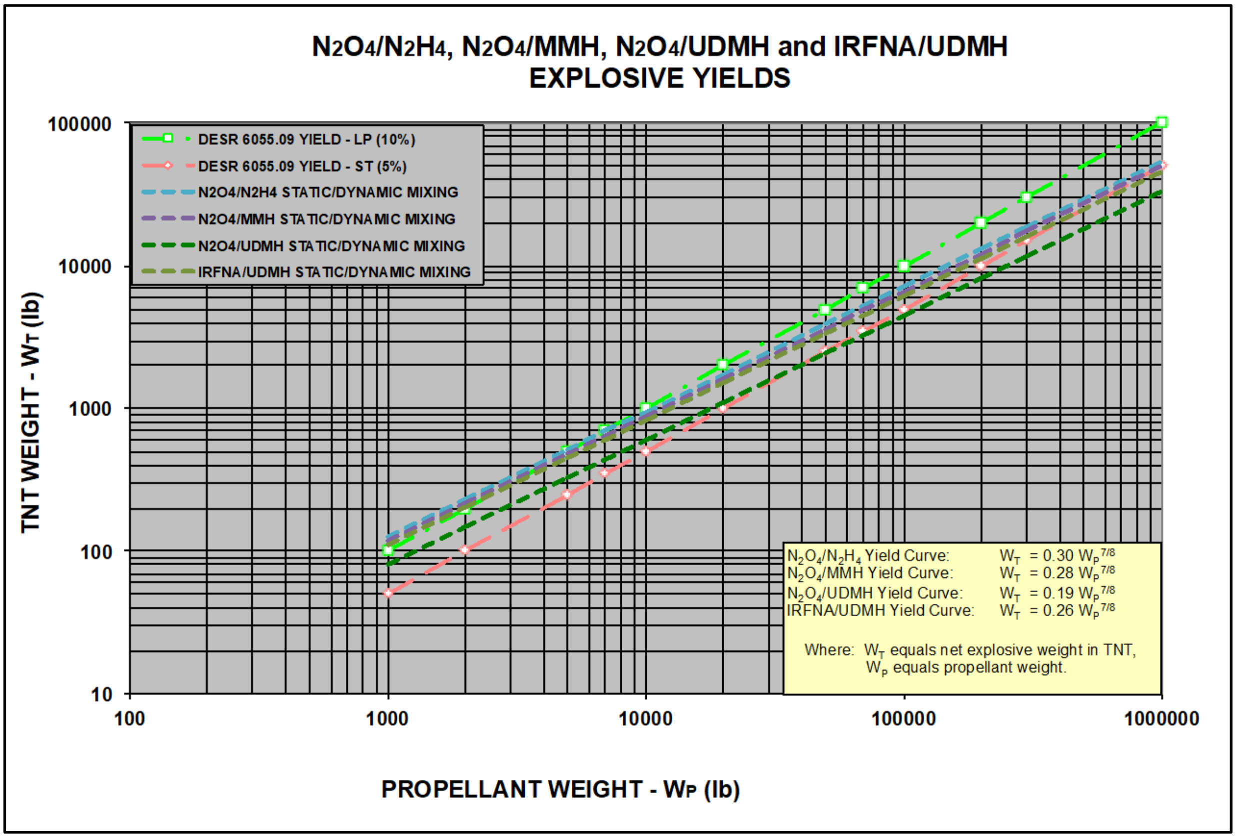


Figure 21. Combined $N_2O_4/N_2H_4$, $N_2O_4$/MMH, $N_2O_4$/UDMH, and IRFNA/UDMH proposed explosive yield curves and current DESR 6055.09 [1] values.

### 5.4.5 Pentaborane (PB) Hypergolic Combination Results

During the late 1950s and 1960s, pentaborane ($B_5H_9$) was investigated as a potential rocket propellant in both the U.S. and the Soviet Union. Pentaborane (PB) is a high-energy, pyrophoric compound that was extensively investigated during development of the missile programs. Applications included both monopropellant and bipropellant jet and rocket engine use. Its highly reactive nature resulted in hypergolic reactions with any oxidizer and high specific impulse performance. Due to its toxicity and acute reactivity, including tendency to react in air, its applicability research was curtailed; however, explosive data were obtained during testing in various combinations and it is appropriate that this information be included here.

The hypergolic bipropellant testing involved combinations of PB with $N_2H_4$, $N_2O_4$, and CTF. Most bipropellant testing was small scale (< 5 lb), although one spill test using 275 lbs of PB and 100 lbs of $N_2H_4$ was attempted. The PB hypergolic reactions differed significantly from the typical $N_2O_4$/hydrazine-based combinations. In the case of PB and either $N_2H_4$ or $N_2O_4$, rather than multiple distributed reactions, the PB combinations exhibited single explosive reactions with reasonable overpressures. Although the PB/$N_2O_4$ tests were not instrumented, the observations indicated that this combination resulted in a greater yield then that of $N_2H_4$/PB. Using similarity and the data in Figure 15 and Figure 16, yield curves were created for both PB/$N_2H_4$ and $N_2O_4$/PB that resulted in $W_T = 0.65\ W_P^{7/8}$ for PB/$N_2H_4$ and $W_T = 1.30\ W_P^{7/8}$ for $N_2O_4$/PB (see Figure 22). Similar to $N_2O_4/N_2H_4$, it is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 40% TNT should be used for PB/$N_2H_4$ and 75% for $N_2O_4$/PB combinations below 1,000 lbs.

Although tested extensively, pentaborane was never used in an operational system. In fact, PB is no longer available; however, due to its past interest as a hypergolic propellant, its data should be retained for the historical record.

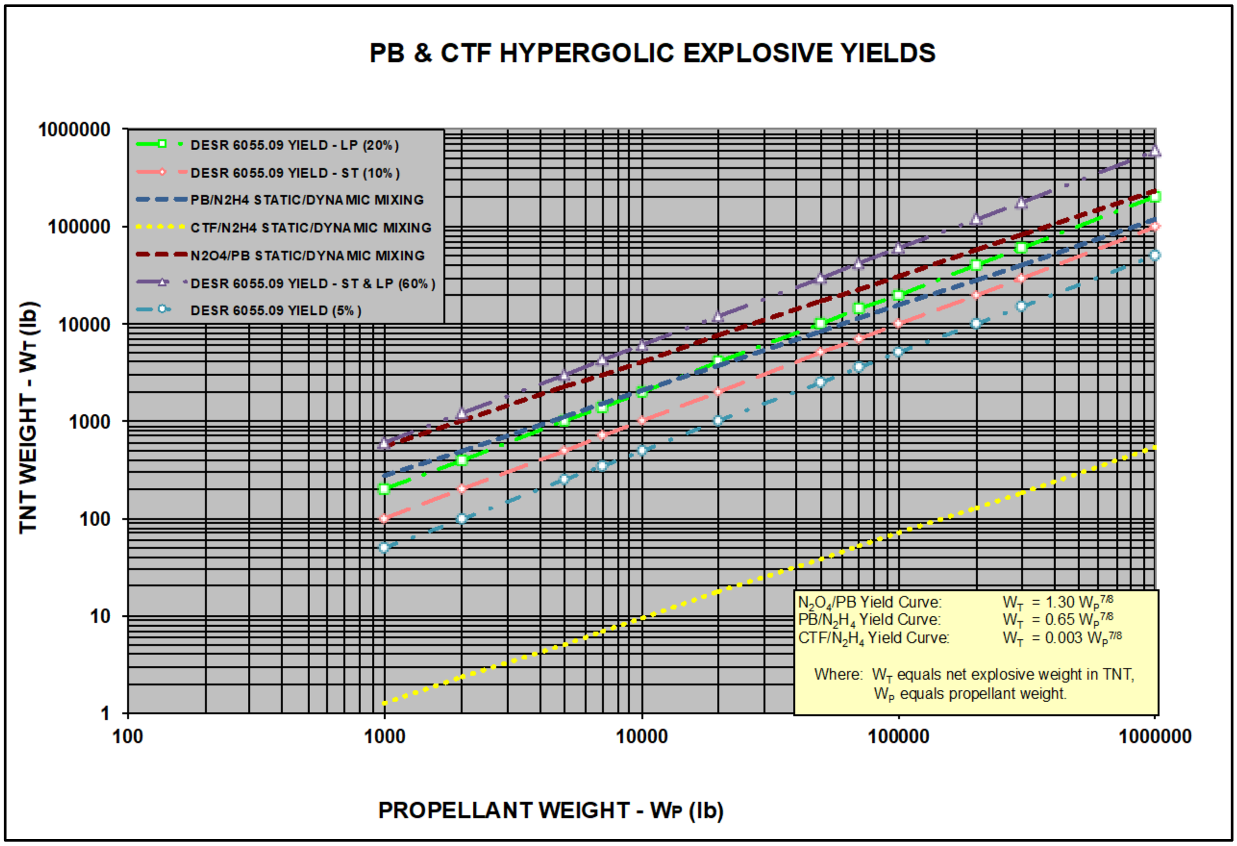


Figure 22. Combined $N_2O_4$/PB, PB/$N_2H_4$, and CTF/$N_2H_4$ proposed explosive yield curves and current DESR 6055.09 values.

### 5.4.6 Chlorine Trifluoride/Pentafluoride (CTF/CPF) Hypergolic Combination Results

CTF ($ClF_3$) is an extremely reactive compound that was pursued as a hypergolic oxidizer for missile systems in the late 1950s. Initially identified in Germany in 1930, use as a rocket propellant began in the late 1940s and early 1950s [75]. Although not adopted in wide use it has been applied to smaller systems and remains an option for packaged systems today.

CTF is a halogenated compound whose hypergolic reactions were found to be the most reactive of any tested but produced the lowest explosive results. This is due to the prompt instantaneous burning encountered when combined, as CTF is reactive with nearly everything but can be stored. CTF's reactivity is surpassed only by elemental fluorine itself. Although toxic and difficult to handle, CTF propellant research continued well into the 1960s.

The CTF hypergolic bipropellant testing involved combinations of $ClF_3$ with $N_2H_4$, and $B_5H_9$. Most bipropellant testing was small scale (< 5 lb), although one spill test using 500 lb of $ClF_3$ and 500 lb of $N_2H_4$ was performed. The CTF hypergolic reactions differed significantly from the typical $N_2O_4$/hydrazine-based combinations. In the case of CTF and either $N_2H_4$ or PB, rather than multiple distributed reactions, the CTF combinations exhibited prompt burning reactions with minimal overpressures. Contrary to expectations, the data indicated that the CTF/$N_2H_4$ combination resulted in a slightly greater yield then that of CTF/PB. Using similarity and the data in Figure 15 and Figure 16, a common yield curve was created for CTF/$N_2H_4$ that resulted in $W_T = 0.003\ W_P^{7/8}$ (see Figure 22). Similar to $N_2O_4$/$N_2H_4$, it is proposed that this value be used for both static and dynamic cases. In terms of explosive potential, however, it is proposed that a maximum of 1% TNT could be used for CTF/$N_2H_4$ below 1,000 lb. An alternative approach could use the individual commodity criteria for CTF and hydrazine, respectively. CTF continues to be used for various purposes and considering past applications of CTF/$N_2H_4$ as a hypergolic propellant in missile systems, this data should be retained, as future use is possible.

Chlorine pentafluoride ($ClF_5$) was discovered by accident in the early 1960s. It was originally identified as “Compound A”: one of two unidentified compounds encountered during chemical experiments. Later identified as $ClF_5$, it retained the Compound A name for many years as a classification measure. Once identified, however, its potential as a hypergolic oxidizer was obvious and production research continued. Although its hypergolic reaction and other attributes are similar to that of $ClF_3$, it results in approximately 20 sec $I_{sp}$ improvement in engine performance [75]. Therefore, although no explosive test data have been found involving CPF, its reaction physics appear to duplicate that of CTF, meaning the CTF/$N_2H_4$ results noted above likely apply to CPF/$N_2H_4$ as well.

# 6. Findings and Recommendations

The findings and recommendations discussed in this section encompass the results of both the prior analyses along with the specific results of this latest hypergolic propellant assessment.

## 6.1 Findings

- Although explosions produced by liquid propellants are significantly different from TNT, the explosive yield in the far-field approximates that of TNT (see Figure 27)
  - Therefore, TNT equivalence is an acceptable criterion for test stand and launch pad siting
- Hypergolic liquid propellant yields are less susceptible to the influence of mixing conditions
- Liquid propellant combination sensitivity to physical environment varies
  - Hypergolic propellants are much different due to their reaction mechanism
- All liquid propellant percent yields decrease as mass increases
  - Therefore, yield curves rather than constant percentage of TNT equivalence is a more appropriate criterion
- Hypergolic reactions inherently display yield asymmetry
- Limited or biased test conditions can result in significant under- or over-prediction of explosive yields under vehicle failure conditions
- Hypergolic static versus dynamic yields differ by a factor of 1.25
- Hypergolic propellant yields are governed by a spontaneous surface area reaction mechanism
- Hypergolic propellants inherently do not represent a significant explosive hazard
  - Surface area generation and contact governs net explosive yield
  - However, this phenomenon should not be generalized for all combinations
    - The presence of $N_2H_4$ in any combination enhances the explosive potential due to its vapor phase reaction
- Liquid propellant combinations tested to date do not act like HD 1.1 explosives
  - Only mixed quantities explode as rapid deflagration
    - Far-field effects are due to significant afterburning
    - Near-field effects analysis requires propellant-specific explosive data
  - Pressure fields do not emulate TNT

- Reaction mass detonation not observed
- Interface generation time determines participation

- Explosive hazards must be validated by comprehensive (distributed, static, and dynamic) test programs
  - Dynamic modes include:
    - Fallback, toppling, impact, donor
  - Static modes include:
    - Tank rupture, inter-tank rupture, line leakage, fire
- Explosive yields of launch vehicles are affected by their configuration and accident sequence
  - For large vehicle quantities, mixing time prior to ignition is the primary contributor to explosive yield

## 6.2 Recommendations

As a result of this assessment, several recommendations and selected options are included regarding the explosive yield values for the hypergolic combinations.

- Recommend replacing the explosive criteria in [1], Table V5.E4.T5 for each hypergolic bipropellant combination as noted
  - Combinations include $N_2O_4$/A-50, $N_2O_4/N_2H_4$, $N_2O_4$/MMH, $N_2O_4$/UDMH, and IRFNA/UDMH
- Recommend retaining explosive data as tentative criteria options for PB and CTF/CPF combinations in a database in DESR 6055.09

### 6.2.1 Recommendation for $N_2O_4$/A-50

| | |
|---|---|
| Launch facility: | $W_T = 0.25\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.20\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 14\%W_P$ |

### 6.2.2 Recommendation for $N_2O_4/N_2H_4$

| | |
|---|---|
| Launch facility: | $W_T = 0.30\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.30\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 18\%W_P$ |

### 6.2.3 Recommendation for $N_2O_4$/MMH

| | |
|---|---|
| Launch facility: | $W_T = 0.28\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.28\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 16\% W_P$ |

### 6.2.4 Recommendation for $N_2O_4$/UDMH

| | |
|---|---|
| Launch facility: | $W_T = 0.19\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.19\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 12\% W_P$ |

### 6.2.5 Recommendation for IRFNA/UDMH

| | |
|---|---|
| Launch facility: | $W_T = 0.26\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.26\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 15\% W_P$ |

### 6.2.6 Option for PB/$N_2H_4$

| | |
|---|---|
| Launch facility: | $W_T = 0.65\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 0.65\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 40\% W_P$ |

### 6.2.7 Option for $N_2O_4$/PB

| | |
|---|---|
| Launch facility: | $W_T = 1.30\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing facility: | $W_T = 1.30\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 75\% W_P$ |

### 6.2.8 Option for CTF/$N_2H_4$ and CPF/$N_2H_4$

| | |
|---|---|
| Launch facility: | $W_T = 0.003\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Test or processing fFacility: | $W_T = 0.003\ W_P^{7/8}$ for all $W_P \geq W_{Pmin}$ |
| Other PES: | $W_T = 1\% W_P$ |

or use individual commodity siting criteria in [1], Table V5.E4.T3.

Where:

$W_T$ = net explosive weight in lb of TNT
$W_P$ = propellant weight in lb as defined in [1], V5.E4.5.6.1, and V5.E4.5.6.2
$W_{Pmin}$ = 1,000 lb is the minimum value applicable to the yield curves
MON can be substituted for $N_2O_4$

## 7. Acronyms and Abbreviations

| | |
|---|---|
| A-50 | Aerozine-50 |
| BoM | Bureau of Mines |
| CBGS | confined-by-ground surface |
| CBM | confined-by-missile |
| cc | cubic centimeter |
| CPF | chlorine pentaflouride |
| CTF | chlorine triflouride |
| DDESB | DOD Explosive Safety Board |
| DESR | Defense Explosives Safety Regulation 6055.09 |
| DMT | distributed mixing test |
| DoD | Department of Defense |
| DoI | Department of Interior |
| EEW | equivalent explosive weight |
| ELWG | Energetic Liquids Working Group |
| ft | feet |
| g | gram(s) |
| HC | hydrocarbon |
| HD | hazard class/division |
| HOVI | hydrogen/oxygen vertical impact |
| HVI | high-velocity impact |
| IBD | inhabited building distance |
| IRFNA | inhibited red fuming nitric acid |
| $I_{sp}$ | specific impulse |
| IWFNA | inhibited white fuming nitric acid |
| lb | pound(s) |
| L/D | length/diameter |
| LP | launch pad |
| LPWG | Liquid Propellant Working Group |
| LSHOE | large-scale hydrogen/oxygen explosion |
| MAF | mixed amine fuel |
| MCE | maximum credible event |
| ml | milliliter |
| MMH | monomethylhydrazine |
| MON | mixed oxides of nitrogen |
| msec | millisecond(s) |
| MSFC | Marshall Space Flight Center |
| NASA | National Aeronautics and Space Administration |

| | |
|---|---|
| NEW | net explosive weight |
| PB | pentaborane |
| PES | potential explosive site |
| PRC | Propellant Reaction Characterization |
| PTRD | public transportation route distance |
| RCS | reaction control system |
| RFNA | red fuming nitric acid |
| SDMH | symmetrical dimethyl hydrazine |
| sec | second(s) |
| SFNA | stable fuming nitric acid |
| ST | static test (stand) |
| TBS | to be supplied |
| TNT | trinitrotoluene |
| UDMH | unsymmetrical dimethyl hydrazine |
| UoF | University of Florida |
| U.S. | United States |
| $W_T$ | TNT weight |
| WFNA | white fuming nitric acid |
| Wp | propellant weight |
| WSTF | White Sands Test Facility |

# 8. References


[1] DESR 6055.09, Ed. 1, Vol. 5, DoD Ammunition and Explosives Safety Standards: Quantity-Distance Criteria for Intentional Burns, Energetic Liquids, and Underground Storage, January 13, 2019.

[2] Tomei, E. J., and J. T. Nichols, *Explosive Equivalence of Hydrocarbon Propellants*, Aerospace Report Number TR-2019-00959, The Aerospace Corporation, El Segundo, CA, April 15, 2019.

[3] Tomei, E. J., "Explosive Equivalence of Cryogenic Propellants," DDESB LPWG, The Aerospace Corporation, El Segundo, CA, June 25, 1997.

[4] Tomei, E. J., "Hypergolic Propellant Hazards in the EPF," The Aerospace Corporation, El Segundo, CA, March 6, 2009.

[5] Tomei, E. J., "Updating the Explosive Equivalence of Liquid Propellants," *28th United States DoD Explosive Safety Seminar*, Orlando, FL, August 19, 1998.

[6] Tomei, E. J., *Propellant Explosives Hazard Study*, Aerospace Report Number TOR-0089(4025-04)-1, Vols. IA, II, III, The Aerospace Corporation, El Segundo, CA, October 1989. Addendum 1, October 2019.

[7] DOD 6055.9-STD, "DoD Ammunition and Explosives Safety Standards," July 1984.

[8] Napadensky, H. S., State-of-the-Art Review of Hypergolic Propellant Explosions, AFATL-TR-89-57, General Research Corp., Santa Barbara, CA, October 1989.

[9] Oslake, J. J.; S. Dobrin; R. J. Getz; N. L. Haight; L. J. Oberste; and R. A. Romine, Launch Siting Criteria for High Thrust Vehicles, U-108:118, Aeroneutronic Division of Ford Motor Company, Newport Beach, CA, March 1961.

[10] Moran, H. E., and R. L. Tuve, Liquid Oxygen – Alcohol Explosion Tests at the Naval Powder Factory, Indian Head, Maryland, Results of, NRL Restricted Report 3250-53/49 amm, April 5, 1949. [TBS]

[11] Scott, F. E.; J. J. Burns; and B. Lewis, Explosive Properties of Hydrazine, R. I. 4460, Department of the Interior Bureau of Mines, Explosives Branch, Pittsburgh, PA, May 1949.

[12] Tuve, R. L., and H. E. Moran, Liquid Oxygen – Alcohol Mixtures, Results of Study of Detonation Mechanism of; NRL Problem No. 32C07-04D (BuShips Project Order A66-863/49), Interim Report on, NRL Restricted Report 3250-73/49 amm, June 1, 1949. [TBS]

[13] Moran, H. E., and A. W. Bertschy, Liquid Oxygen – Alcohol Explosion Tests at the Naval Powder Factory, Indian Head, Maryland, (Second Series) and Miscellaneous Laboratory Tests, Results of, NRL Restricted Report 3250-93/49 amm, August 3, 1949. [TBS]

[14] Fiock, E. F., Detonation of Alcohol and Oxygen in the Liquid Phase, National Bureau of Standards, July 25, 1949. [TBS]

[15] Perls, T. A., Results Obtained by David Taylor Model Basin During Operation Pushover No 2, Report C-288, The David W. Taylor Model Basin, Navy Department, Washington D. C., December 1949.

[16] "Investigation of Liquid Rocket Propellants," Report No. 820-29, Aerojet General Corp, July 1952. [TBS]

[17] Kallis, F., Report - Investigation of Blast Effects of Various Fuel Gels Formed in Liquid Oxygen, ETM #55-55, Rocketdyne Propulsion Field Laboratory, Santa Susana, CA, March 23, 1955.

[18] Kallis F., and F. Thompson, Report - Investigation of Blast Effects of Fuel-LOX Gel Explosions, ETM #56-19, Rocketdyne Propulsion Field Laboratory, Santa Susana, CA, June 7, 1956.

[19] Jupiter Missile Casualty Study, UERD 20-56, U.S. Navy/Norfolk Naval Shipyard, Underwater Explosive Research Division, Norfolk, WV, December 1956.

[20] Rudder, E., REDSTONE Center Section – JUPITER Propellants Destruction Test, Technical Note ORDAB – DSN No. 42, Structures and Mechanics Laboratory, Army Ballistic Missile Agency, Redstone Arsenal, AL, December 26, 1956. [TBS]

[21] Potential Explosive Yields During Static Testing of Missiles and Missile Configurations Employing Liquid Oxygen and Jet Fuel, ABMA IOD 2435-58-1, Combustion and Explosives Research Inc., Redstone Arsenal, AL, October 1958. [TBS]

[22] Explosive Potential of Atlas Propellants, BRC-57-6AI, Broadview Research Corporation, Burlingame, CA, June 1957.

[23] 1/18 Scale Model Missile Tests, Phase IV, Frangible Silo Tests, RD-58-2.1039-10, Broadview Research Corporation, Burlingame, CA, October 1958.

[24] 1/18 and 1/10 Scale Model Missile Tests, Phase V, Flat Pad Tests, RD-58-3.1039-11, Broadview Research Corporation, Burlingame, CA, November 1958.

[25] Smalley, W. M., and D. E. Anderson, The Explosive Potential of Liquid Oxygen and RP-l Missiles, GM-TR-59-0000-00579, Space Technology Laboratories, Los Angeles, CA, January 1959.

[26] Yee, T. S. H., Explosive Equivalence of Liquid Oxygen-RP Mixtures to TNT, R-1476, Rocketdyne Division of North American Aviation Inc., Canoga Park, CA, May 7, 1959.

[27] Final Report on an Investigation of Hazards Associated with the Storage and Handling of Liquid Hydrogen, AF18(600)-1687, Arthur D. Little, Inc., Cambridge, MA, March 1960.

[28] Research on the Fire and Explosion Hazards Associated with New Liquid Propellants, Annual Report, U.S. Bureau of Mines, Department of Interior, April 1960. [TBS]

[29] Final Report on an Investigation of Hazards Associated with the Storage and Handling of Liquid Hydrogen in Close Proximity to Liquid Oxygen and RP-l, AF18(600)-1687, Arthur D. Little, Inc., Cambridge, MA, July 1960.

[30] Research on the Fire and Explosion Hazards Associated with New Liquid Propellants, Progress Reports No. 1 through No. 7, U.S. Bureau of Mines, Department of Interior, October 1960. [TBS] (January 1961)

[31] Review of Fire and Explosion Hazards of Flight Vehicle Combustibles, U.S. Bureau of Mines, Department of Interior, November 1960. [TBS]

[32] Flammability Characteristics of Hydrazine-Unsymmetrical Dimethylhydrazine-Nitrogen Tetroxide-Air Mixtures, Final Report No. 3806, U.S. Bureau of Mines, Department of Interior, February 15, 1961. [TBS]

[33] Research on Hazard Classification of New Liquid Rocket Propellants, Final Report, Vol I, AF/SSD-TR-61-40, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA, October 1961.

[34] Research on Hazard Classification of New Liquid Rocket Propellants, Final Report, Vol II, Titan II Model Missile Tests, AF/SSD-TR-61-40, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA, October 1961.

[35] Willoughby, A. B.; C. Wilton; and T. C. Goodale, A Study of Atlas Missile Failure in Silos, BRC 174-2, Broadview Research Corporation, Burlingame, CA, October 1961.

[36] Flammability Characteristics of Hydrazine-Unsymmetrical Dimethylhydrazine-Nitrogen Tetroxide-Air Mixtures, Final Report No. 3844, U.S. Bureau of Mines, Department of Interior, December 12, 1961. [TBS]

[37] Review of Fire and Explosion Hazards of Flight Vehicle Combustibles, Technical Report 61-278 plus Supplements 1, 2, 3, Aeronautical Systems Division, Wright-Patterson AFB, April 1961 – April 1964. [TBS]

[38] Summary Report on a Study of the Blast Effect of the Saturn Vehicle, C-63850, Arthur D. Little, Inc., Cambridge, MA, February 1962.

[39] Hunter, H., and E. Larsh, Titan II - Dyna-Soar Destruct Test and Analysis Report, ER 12269, Martin Marietta Corporation, Denver, CO, March 1962.

[40] Friedman, R.; W. P. Barnes, Jr.; and M. Markels, Jr., A Study of Explosions Induced by Contact of Hydrazine-Type Fuels with Nitrogen Tetroxide, ASD-TDR-62-685, Atlantic Research Corporation, Air Force Systems Command, Wright-Patterson AFB, OH, September 1962.

[41] Gayle, J. B., and C. H. Blakewood, Destruct Tests on Scale Model Saturn I Booster, NASA TM X-53007, Marshall Space Flight Center, Huntsville, AL, February 1964.

[42] Pesante, R. E.; M. Nishibayashi; D. G. Frutchey; R. D. Erickson; and W. J. Helm, Blast and Fireball Comparison of Cryogenic and Hvpergolic Propellants, CR-69088, Aerojet-General Corporation, Downey, CA, June 1964.

[43] Pesante, R. E.; R. D. Erickson; D. G. Frutchey; and W. J. Helm, Blast and Fireball Comparison of Crvogenic and Hypergolic Propellants with Simulated Tankage, CR-69114, Aerojet-General Corporation, Downey, CA, June 1964.

[44] Irwin, O. R., and J. L. Waddell, Study of Detonation Induced in Solid Propellants by Liquid Propellant Explosions, 0797-0l (09) QP, Aerojet General Corporation, Sacramento, CA, July 1964.

[45] Gayle, J. B.; C. H. Blakewood; J. W. Bramsford; W. H. Swindell; and R. W. High, Preliminary Investigation of Blast Hazards of RP-l/LOX and $LH_2$/LOX Propellant Combinations, NASA TM X-53240, Marshall Space Flight Center, Huntsville, AL, April 1965.

[46] Willoughby, A. B.; J. Mansfield; T.C. Goodale; and C. Wilton, Summary of Existing Information Concerning the Explosive Potential of the Hypergolic Propellant Combination $N_2O_4$/A-50, AFRPL-TR-65-27, URS Corporation, Burlingame, CA, April 1965.

[47] Willoughby, A. B.; C. Wilton; T. Goodale; and J. Mansfield, Study of Liquid Propellant Blast Hazards, AFRPL-TR-65-144, URS Corporation, Burlingame, CA, June 1965.

[48] Pesante, R. E., and M. Nishibayashi, Evaluation of the Blast Parameters and Fireball Characteristics of Liquid Oxygen/Liquid Hydrogen Propellant, CR-65651, Aerojet-General Corporation, Downey, CA, April 1967.

[49] Christos, T.; Y. Miron; H. James; and H. Perlee, "Combustion Characteristics of Condensed-Phase Hydrazine-Type Fuels with Nitrogen Tetroxide," United States Department of the Interior, Bureau of Mines, Pittsburgh, PA, *J. Spacecraft and Rockets*, Vol. 4, No. 9. September 1967.

[50] Willoughby, A. B.; C. Wilton; and J. Mansfield, Liquid Propellant Explosive Hazards, AFRPL-TR-68-92, Vols. 1, 2, 3, URS Corporation, Burlingame, CA, December 1968.

[51] High, R. W., Some Liquid Oxygen/Liquid Hydrogen Explosive Effects in Controlled Failure-Mode Tests, NASA TN D-5382, Manned Spacecraft, Houston, TX, September 1969.

[52] Test Plan, Large-Scale Hydrogen/Oxygen Explosions (LSHOE), TP-WSTF-676, NASA White Sands Test Facility, Las Cruces, NM, April 1993.

[53] NASA White Sands Test Facility, Large Scale Hydrogen/Oxygen Explosion Project, Special Interim Test Report, WSTF 95-28791, NASA White Sands Test Facility, Las Cruces, NM, January 1995.

[54] NASA White Sands Test Facility, Propellant Reaction Characterization Studies, Special Test Data Report, WSTF 94-28611, NASA White Sands Test Facility, Las Cruces, New Mexico (October 1994)

[55] NASA White Sands Test Facility, Propellant Reaction Characterization Studies, Special Test Data Report, WSTF 94-28722, NASA White Sands Test Facility, Las Cruces, New Mexico (November 1994)

[56] NASA White Sands Test Facility, Propellant Reaction Characterization Studies, Special Test Data Report, WSTF 94-28723, NASA White Sands Test Facility, Las Cruces, New Mexico (January 1995)

[57] NASA White Sands Test Facility, Hydrogen/Oxygen Vertical Impact, Draft Test Report, NASA White Sands Test Facility, Las Cruces, NM, May 1997.

[58] Smith, N. H., "Damage Criteria for the Spacing of ICBM Launchers," Memorandum 271.772, The Ramo-Woolridge Corp., Los Angeles, CA, February 14, 1955.

[59] Smalley, W. M., and D. E. Anderson, The Explosive Potential of Liquid Oxygen and RP-1 Missiles, GM-TR-59-0000-00579, Space Technology Laboratories, Los Angeles, CA, January 1959.

[60] Atlas Missile 9C Failure Report, FTA 6182, Convair Astronautics Division, General Dynamics Corporation, San Diego, CA, October 1959.

[61] Smalley, W. M., Revised Overpressure Curves for Liquid Oxygen and RP-l, TDR-594(1430)TN-l, The Aerospace Corporation, El Segundo, CA, March 1961.

[62] Gayle, J. B., Investigation of S-IV All Systems Vehicle Explosion, NASA TN D-563, Marshall Space Flight Center, Huntsville, AL, September 1964.

[63] White, T. N., Abortive Missile Report, Test 205, AF Missile Test Center, Patrick AFB, FL, March 1965.

[64] Launch Hazards Assessment Program, Report on Test 205 from Pad Abort Measuring System, Pan American World Airways, Inc., Patrick AFB, FL, May 1965.

[65] Kite, F. D.; M. Webb; B. E. Bader; and C. N. Golub, Launch Hazards Assessment Program, Report on Atlas/Centaur Abort, SC-RR-65-333, Sandia Laboratories, Albuquerque, NM, October 1965.

[66] Golub, C. N., "Environmental Parameters of an Aborted Launch," *Third Space Congress Proceedings*, March 1966.

[67] Thilges, J. H., "Data from Missile Post-Ignition Aborts," Internal Memo, TRW Corp., Los Angeles, CA, April 1966.

[68] Debus, K. B., Report of Investigation S-IVB-503 Incident, 1-20-67, NASA Marshall Space Flight Center, Huntsville, AL, February 1967.

[69] Farber, E. A.; J. H. Smith; and E. H. Watts, Prediction of Explosive Yield and Other Characteristics of Liquid Rocket Propellant Explosions, Final Report, NASA-CR-137372, University of Florida, Gainesville, FL, June 1973.

[70] Strehlow, R. A., and W. E. Baker, The Characterization and Evaluation of Accidental Explosions, NASA CR-134779, University of Illinois, Urbana, IL, June 1975.

[71] Marshall, J. W. (AFRPL), and L. J. Ullian (ESMC), Investigation of Scenarios and Sequence of Events for the Titan II Complex 374-7 Explosion, December 1980.

[72] Hancock, S. L., "On the Need to Acquire Soviet Launch Accident Data," Foils Engineering, November 5, 2005.

[73] Osipov, V.; C. Muratov; H. Hafiychuk; D. Mathias; S. Lawrence; and M. Werkheiser, "Hazards Induced by Breach of Liquid Rocket Fuel Tanks: Risks of Cryogenic H2-Ox Fluids Explosions," NASA Ames Research Center, Moffett Field, CA, March 5, 2012.

[74] Bunker, R.; M. B. Eck; J. W. Taylor; and S. Hancock, Correlation of Liquid Propellants, NASA Headquarters RTOP, WSTF-TR-0985-001-02, NASA Johnson Space Center, WSTF, Las Cruces, NM, January 23, 2003.

[75] Clark, J. D., Ignition! An Informal History of Liquid Rocket Propellants, Rutgers University Press, New Brunswick, NJ, 1972.

# 9. Additional Bibliography


DIMAZINE (unsymmetrical dimethylhydrazine): Preliminary Data on U-DETA Blend (60% UDMH, 40% DETA), Food Machinery and Chemical Corporation, Chemicals Division, New York, NY (June 1956).

Storage and Handling of DIMAZINE (unsym-DIMETHYLHYDRAZINE), Food Machinery and Chemical Corporation, Inorganic Chemicals Division, New York, NY.

DIMAZINE Unsymmetrical-Dimethylhydrazine (UDMH): Properties, Applications, Reactions, Food Machinery and Chemical Corporation, Inorganic Chemicals Division, New York, NY.

Cassutt, L. H.; F. C. Maddocks; and W. A. Sawyer, A Study of the Hazards in Storage and Handling of Liquid Hydrogen, AF18(600)-1687, A. D. Little Inc., Cambridge, MA (1959).

Hall, C. J., A Committee Study of Blast Potentials at the Saturn Launch Site and A Contractor Study of Blast Forces on Structures, DMM-TR-9-60, Army Ballistic Missile Agency, Redstone Arsenal, AL (February 1960).

Oslake, J. J.; R. J. Getz; R. A. Romine; and K. Soohoo, Explosive Hazards of Rocket Launches, Technical Report No. U-108:89, Aeroneutronic Division of Ford Motor Co., Newport Beach, CA (November 1960).

Joint Air Force-NASA Hazards Analysis Board, Safety and Design Considerations for Static Test and Launch of Large Space Vehicles, Air Force Missile Test Center, Patrick Air Force Base, FL (June 1961).

Debus, K. H., and Maj. Gen. L. I. Davis, Joint Report on Facilities and Resources Required at Launch Site to Support NASA Manned Lunar Landing Program, MT 61-109546 (July 1961).

A Study of Six to Twelve Million Pound Thrust Launch Vehicles, Final Report, TP 64-007 FPO 61-7, Vols. I, II, III, Lockheed-Georgia Division of Lockheed Aircraft Corp., Marietta, GA (September 1961).

Mechanical Systems Design-Criteria Manual for Pentaborane, AF/SSD-TR-61-3, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Mechanical Systems Design-Criteria Manual for Chlorine Trifluoride, AF/SSD-TR-61-4, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Mechanical Systems Design-Criteria Manual for Nitrogen Tetroxide, AF/SSD-TR-61-5, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Mechanical Systems Design-Criteria Manual for Hydrazine, AF/SSD-TR-61-6, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Hydrazine Handling Manual, AF/SSD-TR-61-7, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Nitrogen Tetroxide Handling Manual, AF/SSD-TR-61-8, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Chlorine Trifluoride Handling Manual, AF/SSD-TR-61-9, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Pentaborane Handling Manual, AF/SSD-TR-61-10, Rocketdyne Division of North American Aviation, Inc., Canoga Park, CA (September 1961).

Zabetakis, M. G., and D. S. Burgess, Research on the Hazards Associated with the Production and Handling of Liquid Hydrogen, Report 5707, Bureau of Mines (1961).

Summary Report on a Study of the Blast Effect of the Saturn Vehicle, C-63850, Arthur D. Little, Inc., Cambridge, MA (February 1962).

Storable Liquid Propellants: Nitrogen Tetroxide/Aerozine 50, Report No. LRP 198, Second Edition, Aerojet-General Corporation, Liquid Rocket Plant, Sacramento, CA (June 1962).

Cook, M. A., and L. L. Udy, Detonation Pressure of Liquid Hydrogen/Oxygen, Intermountain Research and Engineering Co., Salt Lake City, UT (September 1962).

Titan II Storable Propellant Handbook, AFBSD-TR-62-2, Revision B, Bell Aerosystems Company, Buffalo, NY (March 1963).

DoD 4145-21, Quantity Distance Storage Criteria for Liquid Propellants, (March 1964).

Kingery, C. N., and B. F. Pannill, Peak Overpressure vs. Scaled Distance for TNT Surface Bursts (Hemispherical Charges), Memorandum Report No. 1518, Ballistic Research Laboratories, Aberdeen Proving Ground, MD (April 1964).

Gayle, J. B., Liquid Propellant Blast Hazards, CPIA Publication 56, pp 593-600, Chemical Propulsion Information Agency, The Johns Hopkins University, Columbia, MD (August 1964).

Carter, P. B., A Method of Evaluating Blast Parameters Resulting from Detonation of Rocket Propellants, AEDC-TDR-64-200, Arnold Engineering Development Center, Arnold Air Force Station, TN (October 1964).

Bracco, F. V., A Method for Predicting the Air Blast Parameters from Liquid Propellant Rocket Explosions, WR 64-1, Wyle Laboratories, Huntsville, AL (January 1965).

Gayle, J. B., and J. W. Bramsford, Size and Duration of Fireballs from Propellant Explosions, NASA TM X-53314, Marshall Space Flight Center, Huntsville, AL (August 1965).

Bracco, F. V., Air Shock Parameters and Design Criteria for Rocket Explosions, WR 65-21, Wyle Laboratories, Huntsville, AL (September 1965).

Bracco, F. V., "Air Blast Parameters Close to a Liquid Propellant Explosion," Interagency Chemical Rocket Propulsion Group, Hazards Working Group, 2nd Meeting Bulletin, Chemical Propulsion Information Agency, The Johns Hopkins University, Columbia, MD (December 1965).

Willoughby, A. B., "Prepared Comments on Paper by F. V. Bracco Entitled 'Air Blast Parameters Close to a Liquid Propellant Explosion'," Interagency Chemical Rocket Propulsion Group, Hazards Working Group, 2nd Meeting Bulletin, Chemical Propulsion Information Agency, The Johns Hopkins University, Columbia, MD (December 1965).

Van Nice, L. J., and H. J. Carpenter, Thermal Radiation From Saturn Fireballs, 2122-6001-T0000, Vol. 1, TRW Space Technology Laboratories, Redondo Beach, CA (December 1965).

Bracco, F. V., Air Blast Parameters Close to a Liquid Propellant Explosion, WR 66-3, Wyle Laboratories, Huntsville, AL (January 1966).

Farber, E. A., "A Mathematical Model for Defining Explosive Yield and Mixing Probabilities of Liquid Propellants," Third Space Congress Proceedings (March 1966).

Farber, E. A., and J. H. Deese, "A Systematic Approach for the Analytical Analysis and Prediction of the Yield from Liquid Propellant Explosions," Third Space Congress Proceedings (March 1966).

Barker, L. I., *Titan IIID Prelaunch Explosion Hazard Analysis*, Aerospace Report Number TOR-669(6124)-4, The Aerospace Corporation, El Segundo, CA (July 1966).

Kingery, C. N., Air Blast Parameters versus Distance for Hemispherical TNT Surface Bursts, Report No. 1344, Ballistic Research Laboratories, Aberdeen Proving Ground, MD (September 1966).

Kite, F. D., and B. E. Bader, Pad-Abort Thermal Flux Model for Liquid Rocket Propellants, SC-RR-66-577, Sandia Laboratories, Albuquerque, NM (November 1966).

Eshelman, C. R., Explosive Phenomena Charts for Missile Safety Determinations, OA Project 67-6, Office of Operations Analysis, 1STRAD, Vandenberg Air Force Base, CA (February 1967).

Holzmann, R. T., ed, Production of the Boranes and Related Research, Academic Press, Inc., New York, NY (1967).

Kennedy, P. E., *Evaluation of Explosive Hazard Criteria and Safety Practices Associated with Titan III Launch Facility Siting. Design and Operations*, Aerospace Report Number TOR-0158(3302)-1, The Aerospace Corporation, El Segundo, CA (January 1968).

Hydrogen Safety Manual, NASA TM X-52454, NASA Lewis Research Center, Cleveland, OH (1968).

KSC Explosives Safety Handbook, GP-469, NASA Kennedy Space Center, FL (July 1968).

Farber, E. A.; F. W. Klement; and C.F. Bonzon, Prediction of Explosive Yield and Other Characteristics of Liquid Propellant Rocket Explosions, University of Florida, Gainesville, FL (October 1968).

Fletcher, R. F., "Characteristics of Liquid Propellant Explosions," NY Academy of Sciences (October 1968).

Monger, J. M., Storable Concentrated Hydrogen Peroxide, AFRPL-TR-68-228, Shell development Company, Emeryville, CA (December 1968).

Mansfield, J. A., Heat Transfer Hazards of Liquid Propellant Explosions, AFRPL-TR-69-89, URS Corporation, Burlingame, CA (February 1969).

Grubbs, B. R.; H. G. Schaeffer; and W. E. Baker, Impact Design Criteria for Blockhouses Subjected to Abortive Launch Environments, PWR-TP-70-2, Mechanics Research Inc., Houston, TX, (March 1969).

Richey, C. M., Project PYRO Dynamic Pressure Accuracy Evaluation, AFRPL-TR-68-111, AF Rocket Propulsion Laboratory, Edwards AFB, CA (June 1969).

Gunther, P., and G R. Anderson, Statistical Analysis of Project PYRO Liquid Propellant Explosion Data, TM-69-1033-3, Bellcomm Inc., Washington, D.C. (July 1969).

USAF Propellant Handbooks, Volume I: Hydrazine Fuels, AFRPL-TR-69-149, Bell Aerospace Company, Buffalo, NY (March 1970).

USAF Propellant Handbooks, Volume II: Nitric Acid/Nitrogen Tetroxide Oxidizers, AFRPL-TR-76-76, Martin Marietta Corp, Denver, CO (February 1977).

Bader, B. E.; A. B. Donaldson; and H. C. Hardee, Liquid-Propellant Rocket Abort Fire Model, SC-RR-70-454, Sandia Laboratories, Albuquerque, NM (October 1970).

Farber, E. A., Critical Mass (Hypothesis and Verification) of Rocket Propellants, Report No. IX, University of Florida, Gainesville, FL (September 1971).

*TNT Equivalency Study for Space Shuttle (EOS)*, Aerospace Report Number ATR-71(7233)-4, Vols. I, II, III, The Aerospace Corporation, El Segundo, CA (September 1971).

Chemical Rocket/Propellant Hazards, CPIA Publication 194, Vols. I, II, III, Chemical Propulsion Information Agency, The Johns Hopkins University, Columbia, MD (October 1971).

Bader, B. E.; A. B. Donaldson; and H. C. Hardee, "Liquid Propellant Rocket Abort Fire Model," Journal of Spacecraft and Rockets, Vol. 8, No 12 (December 1971).

Farber, E. A.; J. H. Smith; and E. H. Watts, Electrostatic Charge Generation and Auto-ignition Results of Liquid Rocket Propellant Experiments, Report No. X, University of Florida, Gainesville, FL (October 1972).

Napadensky, H. S.; J. J. Swatosh Jr.; and D. R. Morita, "TNT Equivalency Studies," Minutes of the Fourteenth Explosives Safety Seminar (November 1972).

Reisler, R. E., "Explosive Yield Criteria," Minutes of the Fourteenth Explosives Safety Seminar (November 1972).

AFM 161-30, Chemical Rocket/Propellant Hazards, Vol. II: Liquid Propellants (April 1973).

Baker, W. E.; V. B. Parr; R. L. Bessey; and P. A. Cox, "Assembly and Analysis of Fragmentation Data For Liquid Propellant Vessels," Minutes of the Fifteenth Explosives Safety Seminar, Vol. 2 (September 1973).

Sutherland, L. C., "A Simplified Method for Estimating the Approximate TNT Equivalent from Liquid Propellant Explosions," Minutes of the Fifteenth Explosive Safety Seminar, Vol. 2 (September 1973).

Baker, W. E.; V. B. Parr; and R. L. Bessey, Assembly and Analysis of Fragmentation Data for Liquid Propellant Vessels, NASA CR 134538, Southwest Research Institute, San Antonio, TX (January 1974).

Smith, J. R., and R. M. Couston, *Space Shuttle Explosive Equivalency Study*, Aerospace Report Number ATR-74(7337)-1, Vols. I, II, III, The Aerospace Corporation, El Segundo, CA (March 1974).

Wilton, C., and A. B. Willoughby, Blast Hazards of the Liquid Propellants LCO and $LN_2O$, AFRPL-TR-74-7, URS Research Company, San Mateo, CA (March 1974).

Boggs, W. H., "Self Limiting Explosive Properties of Liquid Hydrogen and Oxygen," NASA Kennedy Space Center, FL (April 1975).

Lester, D.; A. G. Gibbs; and D. L. Lessor, A Study of Liquid Propellant Auto-ignition, Battelle Pacific Northwest Laboratories, Richland, WA (May 1975).

Boggs, W. H., "Auto Ignition - A Liquid Propellant Explosive Potential Limiting Phenomenon," Thirteenth Space Congress Proceedings (April 1976).

Baker, W. E. et al, Workbook for Predicting Pressure Wave and Fragment Effects of Exploding Propellant Tanks and Gas Storage Vessels, NASA CR-134906, Southwest Research Institute, San Antonio, TX (September 1977).

Sutherland, L. C., "Scaling Law for Estimating Liquid Propellant Explosive Yields," Journal of Spacecraft and Rockets, Vol. 15, No. 2 (March–April 1978).

Baker, W. E. et al, Workbook for Estimating Effects of Accidental Explosions in Propellant Ground Handling and Transport Systems, NASA CR-3023, Southwest Research Institute, San Antonio, TX (1978).

Lyman, O. R., The History of the Quantity Distance Tables for Explosive Safety, Memorandum Report ARBRL-MR-02925, Ballistic Research Laboratory, U.S. Army Armament Research and Development Command, Aberdeen Proving Ground, MD (June 1979).

Tancreto, J. E., Risk Quantification - Part 1 Structural Damage: State-of-the-Art, Technical Memorandum, Civil Engineering Laboratory, Naval Construction Battalion Center, Port Hueneme, CA (January 1981).

DoD Explosives Hazard Classification Procedures, Army Technical Bulletin TB 700-2, Navy NAVORD INST 8020.3, Air Force TO 11A-1-47, Defense Logistics, DLAR 8220.1 (March 1981)

Baeker, J. B.; J. M. Haber; and L. L. Philipson, Effects of Blast Overpressures on Humans and Buildings, Technical Report No 83-1453, J. H. Wiggins Co., Redondo Beach, CA (November 1983).

Hazards of Chemical Rockets and Propellants, CPIA Publication 394, Vols. I, II, III, Chemical Propulsion Information Agency, The Johns Hopkins University, Laurel, MD (September 1984).

Kingery, C. N., and G. Bulmash, Airblast Parameters from TNT Spherical Air Burst and Hemispherical Surface Burst, Technical Report ARBRL-TR-02555, Ballistic Research Laboratories, Aberdeen Proving Ground, MD (April 1984).

AFR 127-100, Explosive Safety Standards (January 1985).

DoDM DoD 4145.26-M, DoD Contractor's Safety Manual for Ammunition and Explosives (March 1986).

Petes, J., Handbook of HE Explosion Effects, Defense Nuclear Agency, DASIAC-TN-86-15, Kaman Tempo, Alexandria, VA (April 11, 1986).

Srinivasa, D. S.; A. L. Kuhl; R. Crawford; W. Asano; and C. Mo, LOX/$LH_2$ Explosion Environment Estimates for the STS-Galileo and Ulysses Projects, RDA-TR-203500-003, R&D Associates, Marina del Rey, CA (March 1988).

Hazard Analysis of Commercial Space Transportation, Vols. I, II, III, DoT Office of Commercial Space Transportation, Washington D. C. (May 1988).

Space Shuttle Data for Planetary Mission Radioisotope Thermoelectric Generator (RTG) Safety Analysis, NSTS 08116 Rev. B, NASA Johnson Space Center, Houston, TX (September 1988).

Parker, L. C.; J. D. Watson; J. F. Stephenson; and G. C. Meyer, Commercial Launch Baseline Assessment: Eastern Space and Missile Center, RTI/4028/01-01F, Research Triangle Institute, Center for Space Systems Engineering, Cocoa Beach, FL (September 1988).

Erdahl, D. C.; D. W. Banning; and E. D. Simon, Space Propulsion Hazards Analysis Manual (SPHAM), AFAL-TR-88-096, Vols. I, II, Martin Marietta Corporation, Denver, CO (October 1988).

Liquid Propellant Manual, CPIA/M4, Chemical Propulsion Information Agency, The Johns Hopkins University, Laurel, MD (November 1988).

Ullian, L. J., "Damage Potential from Prelaunch Catastrophic Events on Adjacent Launch Pads," ESMC/SE Memorandum, Eastern Space and Missile Center, Patrick AFB, FL (January 4, 1989).

Eck, M. B., "Hydrogen-Oxygen Explosion Test Strategy," Fairchild Space Company, Germantown, MD (November 1989).

Eck, M. B., "Hydrogen-Oxygen Explosion Analysis," Fairchild Space Company, Germantown, MD (March 1990).

Rice, E. E.; W. A. Riehl; and T. Kellicut, "Explosion Potential Comparison for Future HLLV Concepts," Summary Presentation, Orbital Technologies Corp., Madison, WI (August 1991).

Test Plan, Large-Scale Hydrogen/Oxygen Explosions (LSHOE), TP-WSTF-676, NASA White Sands Test Facility, Las Cruces, NM (April 1993).

AFM 91-201, Explosive Safety Standards (October 1994).

Taylor, J. W., "Preliminary Analysis of LSHOE," John Taylor Enterprises, Los Alamos, NM (1995).

Taylor, J. W., Ignition Probabilities and Peak Shock Overpressures for Centaur ADS and CSDS in Cassini, John Taylor Enterprises, Los Alamos, NM (December 1996).

Taylor, J. W., "Yields of HOVI Shots versus Cloud Diameter," John Taylor Enterprises, Los Alamos, NM (1997).

Cocchiaro, J. E., Fire and Explosion Hazards of Liquid Propellants and Related Materials, CPIA Publication 661, Chemical Propulsion Information Agency, The Johns Hopkins University, Columbia, MD (October 1997).

Tomei, E. J., "Explosive Equivalence of Liquid Propellants," 1998 JANNAF Propellant Development & Characterization Subcommittee and Safety & Environmental Protection Subcommittee Joint Meeting, NASA JSC, Houston, Texas, The Aerospace Corporation, El Segundo, CA (April 22, 1998).

Ward, J. A., Jr., Baseline Launch-Area Risks for Atlas V 401 Launch, RTI Report No. RTI/6762/10-29F, RTI International, Center for Aerospace Technology, Cocoa Beach, FL (September 30, 2001).

Wall, B., and J. A. Ward Jr., Baseline Launch-Area Risks for Delta-IV Medium+(4,2) Launches, RTI Report No. RTI/8360/03-02F, RTI International, Center for Aerospace Technology, Cocoa Beach, FL (May 10, 2002).

Taylor, J. W., "Final Unedited Report of Analysis of LSHOE & HOVI," John Taylor Enterprises, Los Alamos, NM (January 2003).

Frietas, C. J., and S. Chocron, "Explosive-Hazard Analysis for Reusable Launch Vehicles in On-Pad Environments," J. of Spacecraft and Rockets, Vol. 45, No. 4 (July–August 2008).

Hancock, S. L., "Blast and Fragment Estimates for Atlas V 401 Launch Mishaps," (Draft), Foils Engineering (January 4, 2009).

Lambert, R. R., "Liquid Propellant Blast Yields for Delta IV Heavy Vehicles," ACTA, Lompoc, CA (July 2010).

DoD 6055.09-M, DoD Ammunition and Explosives Safety Standard, Vols. 1 through 8 (August 31, 2018).

Tomei, E. J., "Survey of Near-Pad Explosive Failures," The Aerospace Corporation, El Segundo, CA (September 26, 2018).

## Appendix A. Methods and Observations

### A.1 $N_2O_4$/A-50 Explosive Yield Prediction Methods

Few prediction methods have been developed to estimate the yield for hypergolic bipropellants. These methods are:

1. Farber/UoF (1965-1973) - Mixing/Yield Model and Critical Mass Method
2. URS Corp (1968) - Model Based on PYRO Test Data

Prediction methods were compared to 95% upper-limit values from the above data assessment and are shown in Figure 23. These prediction methods give dissimilar results although Farber appears to be a blend of the three PYRO predictions that are independent of weight. In comparison to the 95% upper-limit curves, our $N_2O_4$/A-50 data assessment is in good agreement with the predictions of PYRO around 300,000 lb.

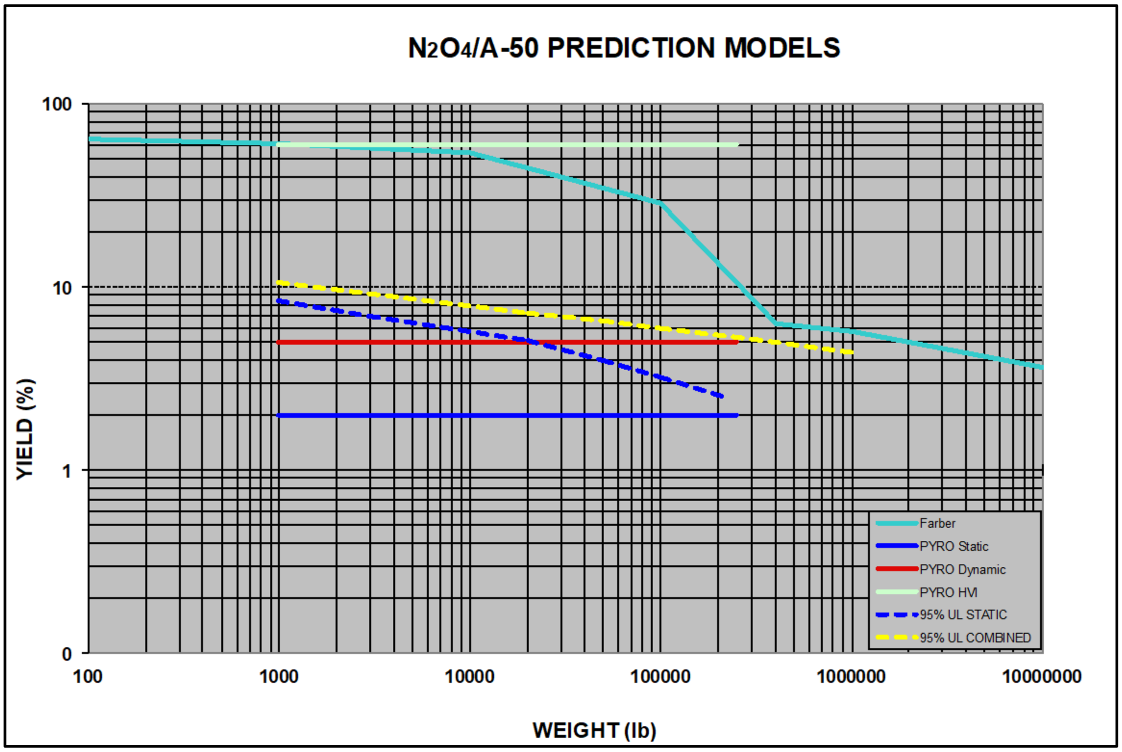


Figure 23. Comparison of $N_2O_4$/A-50 explosive prediction methods.

Figure 24 includes the above prediction methods, our proposed yield curves, the DMT and HVI data, the DESR 6055.09 criteria, and the explosive yield estimate from the Long March CZ-3B failure described in [6], Addendum 1. Figure 24 shows consistency between the Long March failure and the corresponding HVI/DMT data. The HVI and DMT results are not incorporated in our proposed $N_2O_4$/A-50 Yield Curves, as this failure mode does not apply to the DESR criteria. However, our proposed $N_2O_4$/A-50 static and dynamic yield curves approximate the upper criteria (10%) at the lower weights and the lower criteria (5%) at higher weights and are consistent with the shape of the HVI plot.

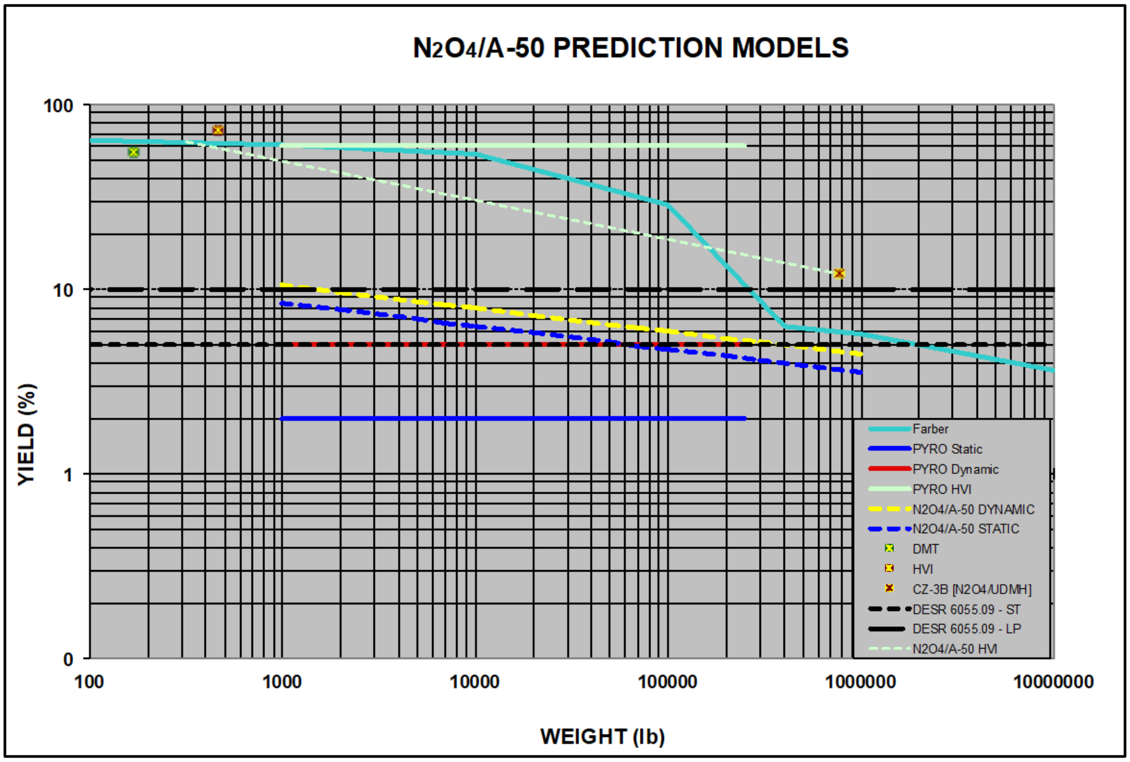


Figure 24. Comparison of $N_2O_4$/A-50 explosive prediction methods with CZ-3B failure.

## A.2 Blast Asymmetry of Hypergolic Propellants

Liquid propellant explosions are often asymmetric and all vehicle failures that have been studied exhibited some degree of asymmetry as the explosive pressure fields exhibit a hemi-ellipsoidal rather than hemispherical form. Review of the test and failure data indicates that this is more prevalent in dynamic mixing modes. This is likely due to a variety of conditions related to launch vehicles such as failure dynamics, rupture and spill mechanics, vehicle orientation, and propellant tank configuration, which are all probable contributing factors. This means that using average yield data (as is done by most experimenters and analysts) underestimates the hazard potential for a given explosion direction, however, using only maximum values risks basing the safety criteria on spurious data. In addition, it was found that the $LO_2/LH_2$ reactions exhibited more severe asymmetry in comparison to $LO_2$/RP-1 due to the inherent differences in the manner in which the propellants are released and mixed.

In the case of these hypergolic mixtures a similar increase in asymmetry was noted due to the lack of a consistent geometry of the reacting products. Therefore, for this assessment, all large-scale ($\geq$ 1,000 lb) test and failure cases were assessed for asymmetry. For $LO_2$/RP-1, the ratio of maximum/average yield was found to range on average from 1.1 to 1.26 where the lowest values correspond to pre-ignition static failure events, while the $LO_2/LH_2$ reactions exhibited as high as a 1.8 factor. Therefore, it was concluded that the higher asymmetry factor should be applied to the results of the hypergolic $N_2O_4$/A-50 combination data assessment in deriving the final recommended yield values. These factors were applied to the 95% upper-limit curves (see Figure 25) and are inherent in the yield curves of all hypergolic combinations based on similarity.

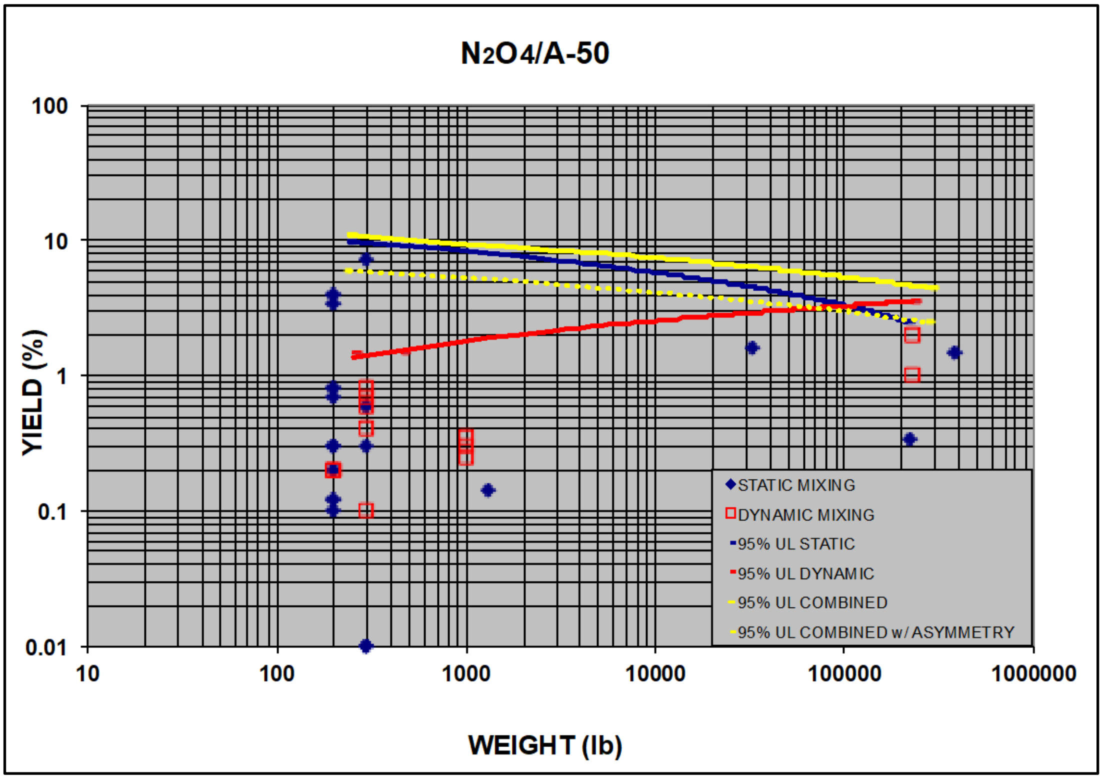


Figure 25. Effect of asymmetry factors on yield estimate.

### A.3 Impact Sensitivity of Hypergolic Propellants

This assessment found a factor of 25% difference in $N_2O_4$/A-50 yields between static and dynamic (≤ 150 ft/sec) conditions. This difference is minimal when compared to the four-times difference of $LO_2$/RP-1. The hypergolic large-scale test and failure data are also very limited compared to the hydrocarbon and cryogenic database. However, a significant influence on yield from high-velocity impact is clearly evident in the data.

This can be readily explained by the inherent nature of the various hypergolic bipropellant combinations. The hypergolic propellants in fact do not mix, as prompt reaction occurs at the interface. As discussed above, the $N_2O_4$/A-50 explosive reaction is a function of the combination of liquid/liquid and vapor droplet products that react spontaneously when they come together. When released separately or at high velocity, larger surface areas and hence reactions can occur (see Figure 26).

The PYRO and Rocketdyne CBGS tests demonstrated that low kinetic energy imparted to the mixing process does not significantly promote increased yield and, in fact, often retards it due to the minimal contact area needed to initiate the reaction sequence. This sequence then results in multiple distributed reactions as the surface contact expands and disperses. This is an inherent probability with any hypergolic combination since the reaction tendency is so pronounced. However, there are optimum conditions of both contact surface generation time and confinement that influence the final reaction yield under high-kinetic-energy conditions.

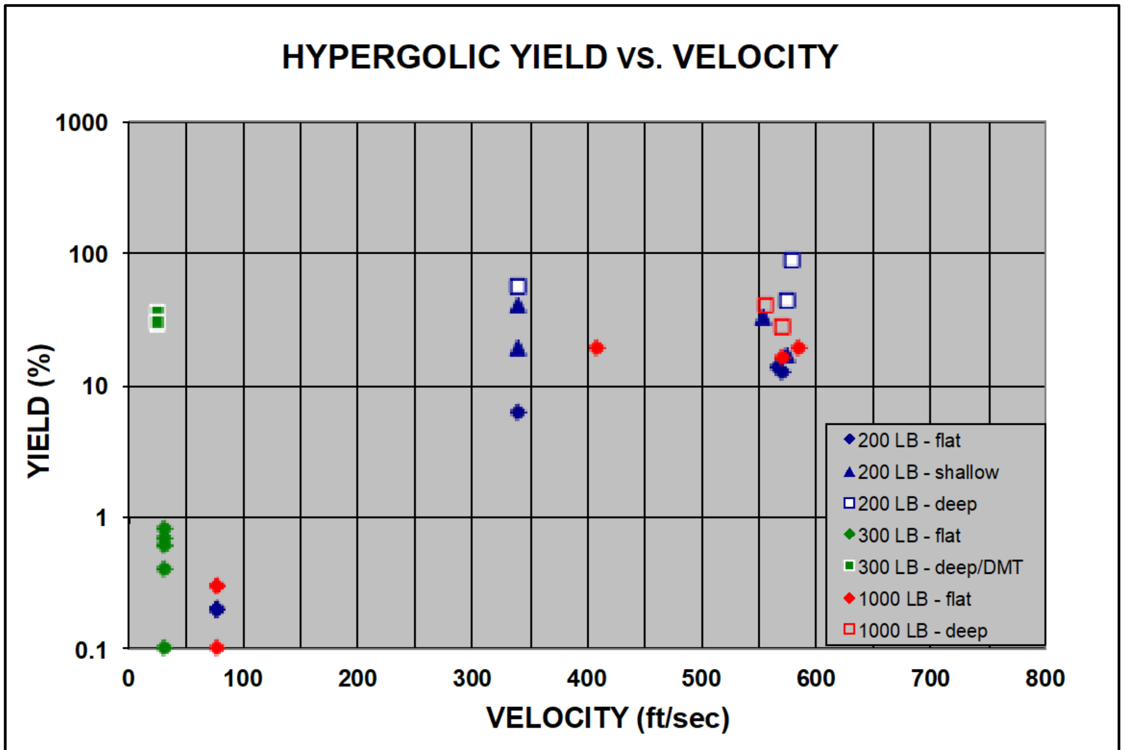


Figure 26. Effect of impact velocity on hypergolic yield.

### A.4 Past Hypergolic Propellant Combinations

There have been literally thousands of hypergolic propellant combinations tested in the laboratories during the 1940s and 1950s beyond the group contained in this report, but explosive test data have not been found for any of them, although detonations were observed in many cases. However, in addition to the hypergolic combinations discussed above ($N_2O_4$/A-50, $N_2O_4/N_2H_4$, $N_2O_4$/MMH, $N_2O_4$/UDMH, IRFNA/UDMH, $B_5H_9/N_2H_4$, $N_2O_4/B_5H_9$, $ClF_3/N_2H_4$ and $ClF_5/N_2H_4$), other combinations have been used operationally in the past in the U.S. These include RFNA/Aniline on the Aerobee, Corporal, WAC Corporal, and Bumper; SFNA/Aniline on Corporal II; IWFNA/UDMH on Vanguard and Thor Able; and IRFNA/JP-X on the SNORT rocket. Although no explosive test data have yet been found (pending search for [16] and others), their explosive attributes are likely no greater than the combinations studied in this report.

### A.5 TNT Equivalence of Liquid Propellants

It has been well established by many analysts—including the authors—that energetic liquids, particularly liquid propellants, do not replicate TNT explosions. For example:

- Near-field overpressures are at least an order of magnitude less than TNT
- Liquid propellant TNT equivalence increases with distance
- Liquid propellant TNT equivalence decreases with increasing propellant weight
- Near-field liquid propellant impulse percent yield is greater than pressure yield
- Liquid propellant impulse yield decreases with distance

However, it has also been shown that these liquid propellant blast environments do approach TNT in the far-field. This is generally attributed to the propensity of the reactants to continue to supply energy to a blast wave for a much longer time than that which influences the initial shock interaction with air. Figure 27, taken from J. Taylor analysis of LSHOE $LO_2/LH_2$ distributed mixing tests (DMT), demonstrates this phenomenon [74]. Although analysts often take issue with the equivalence approach, this is primarily related to the near-field pressure field interactions with other components within the blast environment. However, relative to the use of this method in DESR 6055.09, facility siting using this method is surprisingly appropriate. The difference between the overpressure range of interest for inhabited building (IBD) and public transportation route (PTRD) siting using TNT and liquid propellants is seen to be negligible.

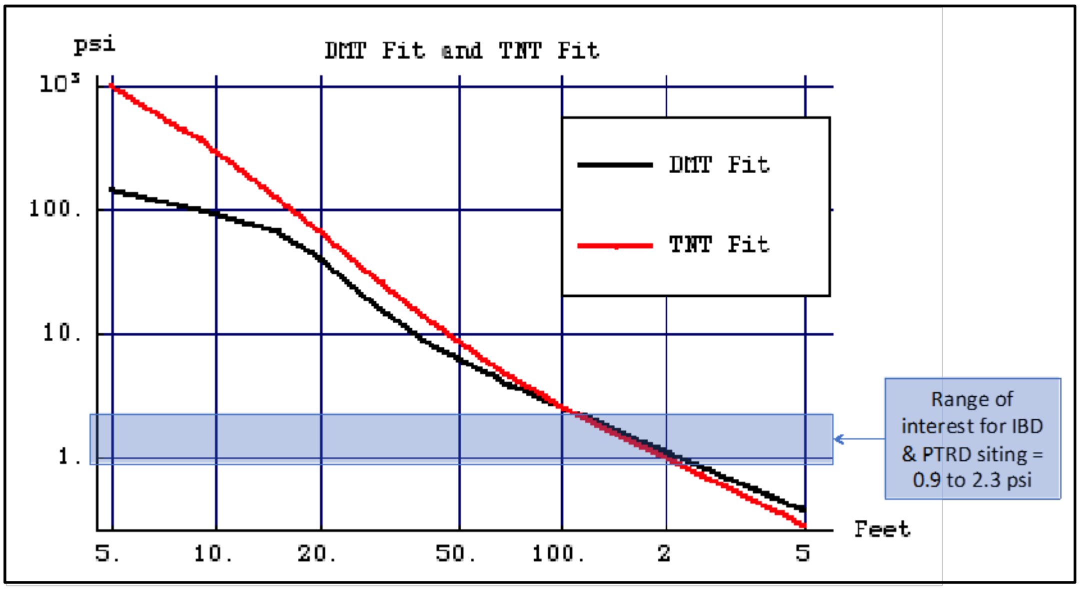


Figure 27. TNT vs. liquid propellant explosive yield [74].

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