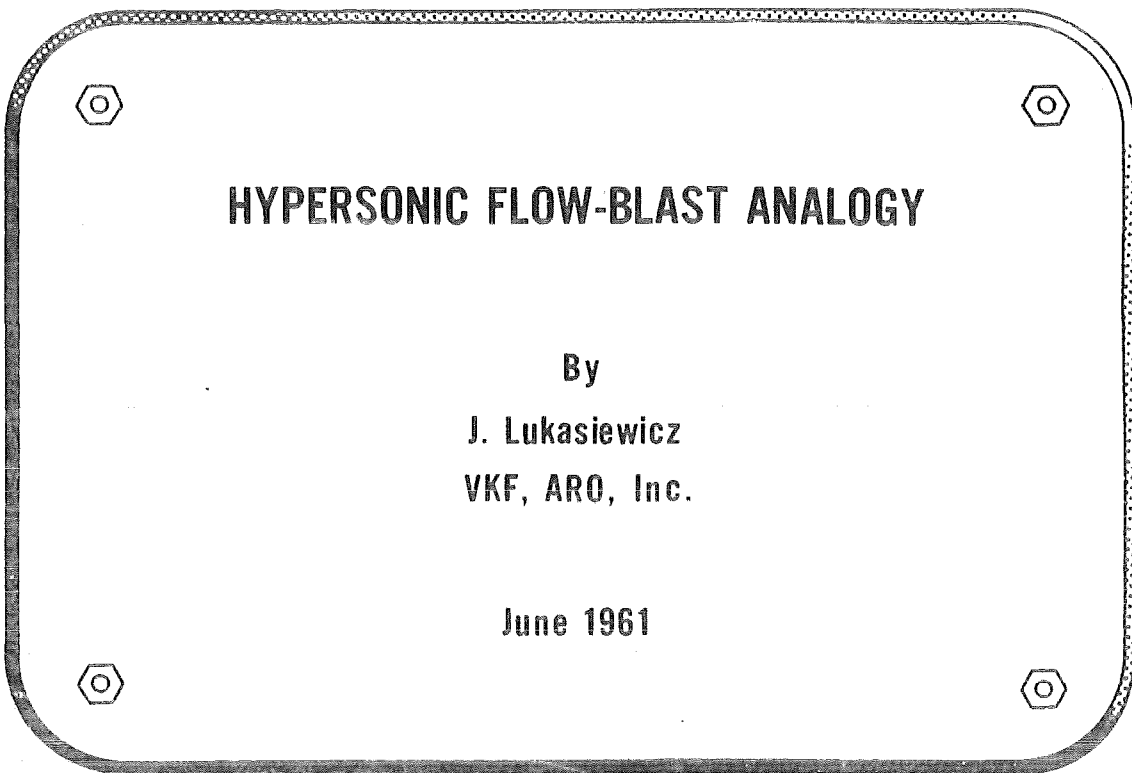
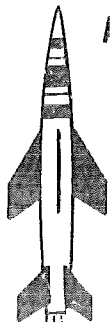


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HYPERSONIC FLOW-BLAST ANALOGY

By

J. Lukasiewicz
VKF, ARO, Inc.

June 1961

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Program Area 750A, Project 8952, Task 89512
ARO Projects 300116, 360105, 360106

Contract No. AF 40(600)-800 S/A 11(60-110)

ABSTRACT

Two-dimensional and axisymmetric, inviscid, hypersonic flows about simple slender bodies are considered with particular reference to pressure distributions and shapes of shock waves. Approximate solutions, based on blast analogy, are derived and compared with more exact, theoretical calculations and with experimental results. The validity of correlation parameters predicted by the blast analogy and the range of their applicability are investigated. Near-exact theoretical results available for real airflow at Mach numbers from 15 to 19 are compared with the perfect gas data.

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NOMENCLATURE

A, B	Functions, Eq. (62)
a	Velocity of sound
a, b	Constants
C	Function, Eq. (64)
C_D	Drag coefficient
D	Drag
d	Plate thickness or cylinder diameter
E_α	Explosion energy per unit area of the surface of shock front when $R = 1$
f, g, h	Functions, Eq. (3)
J_o	Constant
M	Mach number
M_n	Mach number component normal to shock front
M_R	Shock Mach number ($= V/a_\infty$)
p	Static pressure; pressure on plate or cylinder surface
p'_o	Pitot pressure
p_{sh}	Shoulder pressure
p_R	Pressure at shock
R	Distance from shock to origin
R_o	$(E_\alpha/p_\infty)^{1/(a+1)}$
r	Radial or normal co-ordinate
s	Distance measured along shock
t	Time
u	Velocity component in the free-stream flow direction
V	Velocity of blast wave
v	Velocity component normal to the free-stream flow direction or flow velocity caused by explosion
x	Distance from nose in the free-stream flow direction
x'	Distance from shoulder in the free-stream flow direction

y	M_R^{-2}
a	Constant; = 0 for plane flow, = 1 for axisymmetric flow
β, ν	Functions, Eq. (13)
γ	Ratio of specific heats
κ	r/R
ρ	Density
τ	Body thickness parameter
Φ, χ, ψ	Functions, Eq. (8)
χ'	Function, Eq. (76)

SUBSCRIPTS

∞	Free stream
----------	-------------

SUPERSCRIPTS

()'	First approximation
()''	Second approximation

INTRODUCTION

This report deals with the problem of inviscid hypersonic flow of air over simple slender bodies. Two-dimensional and axisymmetric flows are considered with particular reference to pressure distributions and shapes of shock waves. Approximate solutions, based on blast analogy, are derived and compared with more exact, theoretical calculations and with experimental results. The validity of correlation parameters predicted by the blast analogy and the range of their applicability are investigated. Near-exact theoretical results available for real air-flow at Mach numbers from 15 to 19 are compared with the perfect gas data.

The author wishes to thank Messrs. H. W. Ridyard, R. E. Geiger, and the General Electric Company for permission to use their unpublished flow field computations and Mr. Vernon Van Hise of the NASA Langley Research Center for making his original data available. His thanks are also due to Miss P. Mitchell and Mrs. B. Majors for their help with the preparation of this report.

BASIS OF BLAST ANALOGY

The equations of small disturbance theory of steady, hypersonic flow (Ref. 1) may be written as:

Continuity:

$$u_{\infty} \rho_x + (\rho v)_r + a \rho v / r = 0$$

Momentum:

$$u_{\infty} v_x + v v_r + p_r / \rho = 0$$

Energy:

$$u_{\infty} (p \rho^{-\gamma})_x + v (p \rho^{-\gamma})_r = 0$$

(1)

with $a = 0$ for plane and $a = 1$ for axisymmetric flow.

As pointed out by Van Dyke (Ref. 1) and Hayes (Ref. 2), these equations do not involve the streamwise velocity u , and therefore the flow in the transverse planes can be treated independently.

The equations of one-dimensional, unsteady flow are:

Continuity:

$$\rho_t + (\rho v)_r + a\rho v/r = 0$$

Momentum:

$$v_t + vv_r + p_r/\rho = 0$$

Energy:

$$(\rho\rho^{-\gamma})_t + v(\rho\rho^{-\gamma})_r = 0$$

(2)

Hayes (Ref. 2) pointed out that the steady flow, as described by the small disturbance theory, is analogous to the unsteady flow, since, by putting $u_\infty = dx/dt$, Eq. (1) becomes identical with Eq. (2). In other words, for an observer stationary with respect to the undisturbed fluid, the small disturbance theory flow is given by the unsteady solution in one fewer space co-ordinates, so that flow in the plane normal to the direction of motion can be treated independently of x .

If a solution could be obtained for the simpler unsteady problem, then the corresponding steady-flow problem could also be solved; this procedure has been generally known as the blast-wave analogy method. Before this method is applied herein, it is pertinent to recognize the assumptions inherent in the steady and unsteady solutions involved and to examine their compatibility.

The hypersonic small disturbance theory is applicable to slender bodies, such that the body thickness parameter $\tau \ll 1$ (since terms of the order of τ^2 are neglected), and at large Mach numbers, such that $M_\infty \tau \geq 1$. Since the longitudinal velocity component is eliminated in this approximation, the drag of a body is given entirely by the energy of the transverse flow: the drag work expended by moving the body to a given transverse plane is equal to the increment of energy of the transverse flow up to that same plane. The energy of the flow in the transverse plane per unit length will not change when the slope of the body surface is zero; therefore, within the hypersonic, small disturbance approximation, the energy of the transverse flow will remain constant downstream of the forebody or nose when it is followed by a zero-slope afterbody, such as a flat plate or cylinder.

Turning now to the case of unsteady flow, approximate solutions for the spherical, cylindrical, and plane unsteady flows with constant energy have been obtained for cases in which the shock wave which contains the flow field is strong, the shock Mach number being large.

Consider a configuration consisting of a blunt nose followed by a zero-slope, flat plate or cylindrical afterbody at a large flight Mach number. For such a body, the hypersonic small disturbance theory will not apply in the nose region, since $r \ll 1$ is violated in that region. Also, it will not apply far downstream, where the shock has decayed and flow deflections are negligible, because no longer is $M_\infty r \geq 1$. Therefore, for a body of this type, the hypersonic, small disturbance theory will apply in a region of certain limited extent downstream of the nose. In this region, the body surface slope is zero, the energy in the transverse plane is constant, and hence the unsteady solutions are applicable. The success of the blast-wave analogy will thus largely depend on the accuracy of the solutions of unsteady, constant energy flows.

APPROXIMATE SOLUTIONS OF UNSTEADY, PLANE AND CYLINDRICAL, CONSTANT ENERGY FLOW

Following the work of Taylor (Ref. 3), Sakurai (Ref. 4) obtained first and second approximations to unsteady, constant energy, plane, cylindrical, and spherical flows. In this section, Sakurai's results are summarized for use with the blast analogy method, and, in general, his notation is adhered to.

FIRST APPROXIMATION

Approximate blast-wave solutions are constructed in the form of power series in $y = M_R^{-2}$, where M_R is the shock Mach number. In the first approximation, all powers of y except zero are neglected; that is, a strong shock is assumed. The following equations result:

$$\left. \begin{aligned} v/V &= f^{(0)}(\kappa) \\ p/p_\infty &= g^{(0)}(\kappa)/y \\ \rho/\rho_\infty &= h^{(0)}(\kappa) \\ y(R_0/R)^{(\alpha + 1)} &= J_0 \end{aligned} \right\} \quad (3)$$

with $R_0 = (E_\alpha/p_\infty)^{1/(\alpha + 1)}$ where E_α is the explosion energy per unit area of the surface of the shock front when $R = 1$. $f^{(0)}$, $g^{(0)}$, and $h^{(0)}$ are functions of $\kappa = r/R$, γ and α , whereas J_0 is a constant depending on γ and α only.

They are given by:

$$\left. \begin{aligned}
 f^{(0)}(\kappa) &= \kappa/\gamma + A\bar{x}^n \\
 g^{(0)}(\kappa) &= \frac{2\gamma}{\gamma + 1} \left(\frac{\gamma + 1 - \kappa(n - 1)}{\gamma} \right) - \frac{2\gamma^2 + (3\alpha + 1)\gamma - (\alpha + 1)}{(1 - \alpha)\gamma + 3\alpha + 1} \\
 h^{(0)}(\kappa) &= \frac{\gamma + 1}{\gamma - 1} \kappa \frac{\alpha + 1}{\gamma - 1} \left(\frac{\gamma + 1 - \kappa(n - 1)}{\gamma} \right) - \frac{2(2\alpha + 1 + \gamma)}{(1 - \alpha)\gamma + 3\alpha + 1} \\
 \text{with } A &= \frac{1}{\gamma} \frac{\gamma - 1}{\gamma + 1}, \quad n = \frac{(2 - \alpha)\gamma^2 + (3\alpha + 1)\gamma - 1}{\gamma^2 - 1}
 \end{aligned} \right\} \quad (4)$$

The variation of shock position and pressure with time are obtained from Eq. (3) with $V = dR/dt$. Thus

$$\frac{ta_\infty}{R_0} = \sqrt{J_0} \frac{2}{\alpha + 3} \left(\frac{R}{R_0} \right)^{(\alpha + 3)/2} \quad (5)$$

and

$$\frac{ta_\infty}{R_0} = \sqrt{J_0} \frac{2}{\alpha + 3} \left[\left(\frac{g^{(0)}}{J_0} \right) / \left(\frac{p}{p_\infty} \right) \right]^{(\alpha + 3)/[2(\alpha + 1)]} \quad (6)$$

The boundary or shock conditions ($\kappa = 1$) are given, to this approximation, by

$$\frac{v}{V} = \frac{2}{\gamma + 1}, \quad \frac{p}{p_\infty} \frac{1}{M_R^2} = \frac{2\gamma}{\gamma + 1}, \quad \frac{\rho}{\rho_\infty} = \frac{\gamma + 1}{\gamma - 1} \quad (7)$$

The values computed by Sakurai (Ref. 4) at $r = 0$ ($\kappa = 0$) for $\gamma = 1.4$ are given in Table 1. For other values of γ , J_0 is given in Table 2, and other functions can be computed from Eq. (4).

SECOND APPROXIMATION

In the second approximation, all powers of y except zero and one are neglected, and the following equations result:

$$\begin{aligned} v/V &= f^{(0)}(\kappa) + (\kappa - f^{(0)}(\kappa))\Phi(\kappa)y \\ p/p_\infty &= g^{(0)}(\kappa) (1 + \psi(\kappa)y)/y \\ \rho/\rho_\infty &= h^{(0)}(\kappa) (1 + \chi(\kappa)y) \end{aligned} \quad (8)$$

$$y(R_0/R)^{(\alpha + 1)} = J_0 (1 + \lambda_1 y)$$

where Φ , ψ , and χ are functions of κ , γ , α , and λ_1 depends on γ and α .

The shock position is again obtained in terms of time as

$$\frac{ta_\infty}{R_0} = \sqrt{J_0} \int_0^{R/R_0} \left[(R/R_0)^{-(\alpha + 1)} - J_0 \lambda_1 \right]^{-1/2} d(R/R_0) \quad (9)$$

and the pressure ratio in terms of shock location as

$$p/p_\infty = g^{(0)} \left[(R_0/R)^{(\alpha + 1)} / J_0 - (\lambda_1 - \psi) \right] \quad (10)$$

For $R \rightarrow \infty$ or at infinitely large time, Eq. (10) gives at $r = 0$ ($\kappa = 0$)

$$\left. \begin{aligned} p/p_\infty &= -g^{(0)}(\lambda_1 - \psi) \\ &= 0.692 \text{ for } \alpha = 0 \\ &= 0.606 \text{ for } \alpha = 1 \end{aligned} \right\} \gamma = 1.4 \quad (11)$$

For plane wave ($\alpha = 0$), Eq. (9) gives

$$ta_\infty/R_0 = (-\lambda)^{-3/2} \beta/J_0 \quad (12)$$

where

$$\beta = \sqrt{\nu(1 + \nu)} - \log_e \left(\sqrt{1 + \nu} + \sqrt{\nu} \right) \quad (13)$$

and

$$\nu = -J_0 \lambda_1 (R/R_0)$$

For cylindrical wave ($\alpha = 1$),

$$ta_\infty/R_0 = \left\{ 1 - \left[1 - J_0 \lambda_1 (R/R_0)^2 \right]^{1/2} \right\} / \left(\lambda_1 \sqrt{J_0} \right) \quad (14)$$

In this approximation, the exact conditions at the shock are satisfied for velocity and pressure, that is

$$\frac{v}{V} = \frac{2}{\gamma + 1} (1 - y) \tag{15}$$

$$\frac{p}{p_\infty} = \frac{2\gamma}{\gamma + 1} \frac{1}{y} - \frac{\gamma - 1}{\gamma + 1} \tag{16}$$

the above equations corresponding to Eq. (8) with

$$f(1)^{(o)} = \frac{2}{\gamma + 1}$$

and

$$\Phi(1) = - f(1)^{(o)} / (1 - f(1)^{(o)})$$

The exact density ratio across the shock is

$$\frac{\rho}{\rho_\infty} = \left(\frac{\gamma + 1}{\gamma - 1} \right) / \left(1 + \frac{2}{\gamma - 1} y \right) \tag{17}$$

This is not equivalent to the corresponding Eq. (8), except for small y (large M_R), when it is expanded to

$$\frac{\rho}{\rho_\infty} = \left(\frac{\gamma + 1}{\gamma - 1} \right) \left(1 - \frac{2}{\gamma - 1} y \right) \tag{18}$$

with

$$h(1)^{(o)} = \frac{\gamma + 1}{\gamma - 1}$$

and

$$\chi(1) = - \frac{2}{\gamma - 1}$$

The coefficients in the second approximation are given in Table 1 for $r = 0$ ($\kappa = 0$) and $\gamma = 1.4$.

TABLE 1

VALUES OF COEFFICIENTS IN THE FIRST AND SECOND APPROXIMATIONS TO THE BLAST SOLUTION, $\gamma = 1.4$, $\kappa = 0$ (SAKURAI, REFS. 4, 5)

α	$f(0)^{(o)}$	$g(0)^{(o)}$	$h(0)^{(o)}$	J_0	$\Phi(0)$	$\psi(0)$	$\chi(0)$	λ_1
0	0	0.455	0	1.696	-3.86	-0.617	3.86	-2.138
1	0	0.424	0	0.877	-3.5	-0.56	3.5	-1.989

TABLE 2
VALUES OF J_0 (SAKURAI, REF. 4)

$\gamma \backslash \alpha$	0	1
1.2	3.024	1.547
1.3	2.147	1.102
1.667	1.137	0.585

SOME PROPERTIES OF APPROXIMATE SOLUTIONS

In the first approximation, the ratio of pressures at distances κ_1 and κ_2 is constant at all times and is given by (see Eq. (3)),

$$P_{\kappa_1}/P_{\kappa_2} = \frac{g^{(o)}(\kappa_1)}{g^{(o)}(\kappa_2)} \quad (19)$$

For $\gamma = 1.4$, the values obtained for the ratio of pressure at the shock to pressure at the origin are given in Table 3.

TABLE 3
RATIO OF PRESSURE AT SHOCK
TO PRESSURE AT ORIGIN, $\gamma = 1.4$

α	0	1
P_R/P_0	2.56	2.68

The difference in the pressure ratio p/p_∞ , as given by the first (Eq. (3)) and second (Eq. (8)) approximations, is constant at a given κ and γ and is equal to

$$(p/p_\infty)' - (p/p_\infty)'' = \frac{g^{(o)}(\kappa)}{g(\kappa)} \psi(\kappa) > 0 \quad (20)$$

At $\kappa = 0$ and for $\gamma = 1.4$, this amounts to 0.281 ($\alpha = 0$) and 0.237 ($\alpha = 1$).

The relationship between the position of the shock and the shock pressure ratio is given by

$$\left(\frac{R_0}{R}\right)^{\alpha+1} = \frac{J_0}{g^{(o)}(1)} \frac{P_R}{P_\infty} \quad (\text{first approximation}) \quad (21)$$

and
$$\left(\frac{R_o}{R}\right)^{\alpha + 1} = \frac{J_o}{g^{(1)}} \left\{ \frac{p_R}{p_\infty} + [\lambda_1 - \psi(1)] g^{(1)} \right\} \quad (22)$$

(second approximation)

For $\gamma = 1.4$, the second approximation is

$$\left. \begin{aligned} \alpha = 0, (R/R_o) (p_R/p_\infty - 2.33) &= 0.69 \\ \alpha = 1, (R/R_o)^2 (p_R/p_\infty - 2.16) &= 1.33 \end{aligned} \right\} \quad (23)$$

In the first approximation, the numerical term in the bracket is omitted.

The variation of pressure, density, and velocity with distance (κ) from the origin can be judged from Table 4, in which values of coefficients for the first and second approximations are listed. It is apparent that pressure ($g^{(o)}, \psi$) exhibits negligible variation for $\kappa \leq 0.4$. The density variation ($h^{(o)}, \chi$) is also small near the origin, whereas velocity ($f^{(o)}, \Phi$) varies nearly linearly with κ .

TABLE 4
VARIATION OF COEFFICIENTS FOR FIRST AND SECOND APPROXIMATIONS WITH κ ,
FOR PLANE AND CYLINDRICAL FLOW, $\gamma = 1.4$

	α	κ					
		0	0.2	0.4	0.6	0.8	1.0
$g^{(o)}$	0	0.455	0.455	0.461	0.496	0.631	1.167
	1	0.424	0.424	0.425	0.441	0.531	1.167
ψ	0	-0.617	-	-0.573	-0.260	0.451	-0.143
	1	-0.56	-	-	-0.45	0.092	-0.143
$h^{(o)}$	0	0	0.039	0.225	0.669	1.766	6.000
	1	0	0.001	0.019	0.153	0.783	6.000
χ	0	3.86	-	3.633	3.109	1.674	-5.000
	1	3.5	-	-	-	3.002	-5.000
$f^{(o)}$	0	0	0.143	0.287	0.437	0.611	0.833
	1	0	0.143	0.286	0.432	0.597	0.833
Φ	0	-3.86	-	-3.869	-3.803	-3.842	-5.000
	1	-3.5	-	-	-	-3.542	-5.000

APPROXIMATE STEADY-FLOW SOLUTIONS BASED ON BLAST ANALOGY

The steady, blast analogy approximations are obtained from the approximate constant energy, blast solutions by superposition of free-stream velocity $u_\infty = x/t$ and by expression of the energy of unsteady flow in terms of body wave drag. Thus the plane blast solution is analogous to two-dimensional flow over a blunted plate, and the cylindrical blast solution corresponds to the case of axisymmetric flow about a blunted cylinder. In each instance, first and second blast approximations are used to obtain approximate steady-flow solutions.

In obtaining steady-flow solutions, the shock shape, R/d , and the pressure distribution, p_0/p_∞ , on the body surface as a function of distance along the body, x/d , are the main concern. These solutions are strictly applicable only to flat plates and cylinders at zero incidence. The body pressure p_0 , denoted below by p , is taken at $\kappa = 0$, and the body thickness is thus neglected. In view of the invariance of pressure with κ and the small density near the origin (see Table 4) this procedure seems reasonable.

Static pressure and other parameters of flow in the transverse plane are easily obtained from blast solutions, provided the shock shape has been determined. Since

$$M_R = M_\infty (dR/dx) \quad (24)$$

is given from the shock slope, distributions of transverse pressure, density, etc., follow from Eqs. (3) and (8).

It should be noted that, although the second blast-wave approximation gives essentially exact conditions at the shock surface (see Eqs. (15) through (18)), the corresponding blast analogy gives correct values of pressure, etc., at the shock only when the component of Mach number, M_n , normal to the shock, s ,

$$M_n = M_\infty (dR/ds) \quad (25)$$

closely approximates M_R . This, however, is consistent with the assumptions of the hypersonic small disturbance theory.

For the analogous steady-flow solutions, the blast energy, E_α , is expressed in terms of drag. From the definition of E_α ,

$$E_{\alpha=0} = \frac{1}{2} D_{\alpha=0}$$

and

$$E_{\alpha=1} = \frac{1}{2\pi} D_{\alpha=1}$$

where

$D_{\alpha=0}$ = nose drag of plate per unit length of leading edge

and

$D_{\alpha=1}$ = nose drag of cylinder

In terms of drag coefficient, C_D ,

$$\left. \begin{aligned} E_{\alpha=0} &= \gamma \frac{d}{4} p_{\infty} M_{\infty}^2 C_{D\alpha=0} \\ R_{o\alpha=0} &= \gamma \frac{d}{4} M_{\infty}^2 C_{D\alpha=0} \\ E_{\alpha=1} &= \gamma \frac{d^2}{16} p_{\infty} M_{\infty}^2 C_{D\alpha=1} \\ R_{o\alpha=1} &= \sqrt{\gamma} \frac{d}{4} M_{\infty} \sqrt{C_{D\alpha=1}} \end{aligned} \right\} \quad (26)$$

where C_D is based on plate thickness, d , or cylinder frontal area, $\pi d^2/4$, and free-stream dynamic pressure.

FIRST APPROXIMATION

From Eqs. (5) and (6), the following expressions for shock, shock slope, and pressure distribution are obtained.

Plane flow ($\alpha = 0$):

$$\frac{R}{d} = \left(\frac{3}{4}\right)^{2/3} \left(\frac{\gamma}{J_o}\right)^{1/3} C_D^{1/3} \left(\frac{x}{d}\right)^{2/3} \quad (27)$$

$$= 0.774 C_D^{1/3} \left(\frac{x}{d}\right)^{2/3} \quad (28)$$

or

$$\frac{R/d}{M_{\infty}^2 C_D} = 0.774 \frac{(x/d)^{2/3}}{C_D^{2/3} M_{\infty}^2} \quad (29)$$

for $\gamma = 1.4$;

$$\frac{dR}{dx} = \left(\frac{\gamma}{6J_o}\right)^{1/3} C_D^{1/3} \left(\frac{x}{d}\right)^{-1/3} \quad (30)$$

$$= 0.516 C_D^{1/3} \left(\frac{x}{d}\right)^{-1/3} \quad (31)$$

and

$$\frac{p}{p_\infty} = \left(\frac{\gamma}{6J_o} \right)^{2/3} g^{(o)} M_\infty^2 C_D^{2/3} \left(\frac{x}{d} \right)^{-2/3} \quad (32)$$

$$= 0.121 M_\infty^2 \left[C_D / (x/d) \right]^{2/3} \quad (33)$$

for $\gamma = 1.4$.

From Eqs. (29) and (33),

$$p/p_\infty = 0.0936 M_\infty^2 C_D / (R/d) \quad (34)$$

for $\gamma = 1.4$.

Axisymmetric flow ($\alpha = 1$):

$$\frac{R}{d} = \left(\frac{\gamma}{4J_o} \right)^{1/4} C_D^{1/4} (x/d)^{1/2} \quad (35)$$

$$= 0.795 C_D^{1/4} (x/d)^{1/2} \quad (36)$$

or

$$\frac{R/d}{M_\infty \sqrt{C_D}} = 0.795 \sqrt{\frac{x/d}{M_\infty^2 \sqrt{C_D}}} \quad (37)$$

for $\gamma = 1.4$;

$$\frac{dR}{dx} = \frac{1}{2} \left(\frac{\gamma}{4J_o} \right)^{1/4} C_D^{1/4} \left(\frac{x}{d} \right)^{-1/2} \quad (38)$$

$$= 0.397 C_D^{1/4} (x/d)^{-1/2} \quad (39)$$

and

$$\frac{p}{p_\infty} = \frac{1}{8} \sqrt{\frac{\gamma}{J_o}} g^{(o)} M_\infty^2 \sqrt{C_D} / (x/d) \quad (40)$$

$$= 0.067 M_\infty^2 \sqrt{C_D} / (x/d) \quad (41)$$

for $\gamma = 1.4$.

From Eqs. (37) and (41),

$$p/p_\infty = 0.0423 M_\infty^2 C_D / (R/d)^2 \quad (42)$$

for $\gamma = 1.4$.

SECOND APPROXIMATION

From Eqs. (10) to (14), the following expressions for shock shape and pressure distribution are derived:

Plane flow ($\alpha = 0$):

In this case, explicit expressions for shock shape and pressure ratio were not obtained. Thus, for shock shape:

$$\frac{x}{d} = \frac{\gamma}{4J_0} (-\lambda_1)^{-3/2} \beta C_D M_\infty^3 \quad (43)$$

$$= 0.066 \beta C_D M_\infty^3 \quad (44)$$

for $\gamma = 1.4$

where β and ν are given by Eq. (13) and

$$\nu = -(4J_0 \lambda_1 / \gamma) (R/d) / (M_\infty^2 C_D) \quad (45)$$

$$= 10.36 (R/d) / (M_\infty^2 C_D) \quad (46)$$

for $\gamma = 1.4$.

The shock slope is given by

$$\frac{dR}{dx} = \frac{1}{2} \sqrt{\frac{\gamma C_D}{J_0}} \sqrt{\frac{1 + \nu}{R/d}} \quad (47)$$

$$= 0.453 \sqrt{\frac{1 + \nu}{R/d}} \sqrt{C_D} \quad (48)$$

From Eq. (10),

$$\nu = -\lambda_1 / \left[(p/p_\infty) / g^{(o)} + (\lambda_1 - \psi) \right] \quad (49)$$

$$= 0.973 / (p/p_\infty - 0.692) \quad (50)$$

for $\gamma = 1.4$ which, together with Eq. (43) gives pressure distribution.

In the limit when $(t, R, x) \rightarrow \infty$, the pressure ratio is given by Eq. (11) (= 0.692 for $\gamma = 1.4$).

The function β (Eq. (13)), when expressed in terms of pressure ratio p/p_∞ (Eq. (50)), can usually be approximated to better than one percent by an expression of the form

$$\beta = a(p/p_\infty)^b$$

where a and b are given, for $\gamma = 1.4$, in Table 5 for various ranges of p/p_∞ .

TABLE 5
VALUES OF CONSTANTS a AND b FOR VARIOUS
RANGES OF p/p_∞

Range of p/p_∞	a	b
2 - 4	1.30	-1.87
4 - 20	0.87	-1.59
20 - 100	0.69	-1.52

Using this form of β , Equation (43) can be written

$$\frac{x}{d} = \frac{\gamma^a}{4J_0} (-\lambda_1)^{-3/2} C_D M_\infty^3 (p/p_\infty)^b \quad (51)$$

$$= 0.066a C_D M_\infty^3 (p/p_\infty)^b \quad (52)$$

for $\gamma = 1.4$;

or

$$\frac{p}{p_\infty} = \left[\frac{4J_0}{\gamma^a} (-\lambda_1)^{3/2} \frac{x/d}{M_\infty^3 C_D} \right]^{1/b} \quad (53)$$

$$= \left[(15.15/a) \frac{x/d}{M_\infty^3 C_D} \right]^{1/b} \quad (54)$$

for $\gamma = 1.4$

which is analogous to the first approximation, Eq. (32).

A more convenient form of the second approximation for pressure distribution can be derived as follows:

From Eqs. (32), (43), and (49), the pressure ratio $(p/p_\infty)'$, as given by the first approximation at a given x/d , can be expressed in terms of the second approximation pressure ratio $(p/p_\infty)''$ at the same x/d by

$$\left[(p/p_\infty)' = -(2/3)^{2/3} g^{(0)} \lambda_1 \beta^{-2/3} \right]_{x/d} \quad (55)$$

and

$$(p/p_\infty)' = 0.743\beta^{-2/3} \quad (56)$$

for $\gamma = 1.4$

and is therefore a function of $(p/p_\infty)''$ only.

The values of

$$\left[(p/p_\infty)'' , (p/p_\infty)' \right]_{x/d}$$

as obtained from Eq. (56), are tabulated in Table 6 and plotted in Fig. 1 in terms of the difference $(p/p_\infty)'' - (p/p_\infty)'$ vs $(p/p_\infty)''$.

TABLE 6
PRESSURE RATIOS AT SAME x/d
($\alpha = 0, \gamma = 1.4$, FIRST AND SECOND APPROXIMATIONS)

$(p/p_\infty)''_{x/d}$	0.692	0.8	1	2	5	10
$(p/p_\infty)'_{x/d}$	0	0.191	0.428	1.473	4.493	9.503

From Fig. 1, it is evident that the simplified expression

$$(p/p_\infty)'' = (p/p_\infty)' + 0.56 \tag{57}$$

$$= 0.121 M_\infty^2 \left[C_D/(x/d) \right]^{2/3} + 0.56 \tag{58}$$

gives the second approximation pressure ratio to better than 1.5 percent when $p/p_\infty \geq 1, \gamma = 1.4$.

For the shock shape (Eqs. (10) and (26)),

$$\frac{R}{d} = \frac{\gamma}{4J_0} M_\infty^2 C_D / \left[\frac{p/p_\infty}{g^{(o)}} + (\lambda_1 - \psi) \right] \tag{59}$$

$$= M_\infty^2 C_D / \left[10.65(p/p_\infty) - 7.37 \right] \tag{60}$$

for $\gamma = 1.4$, and using the simplified expression (Eq. (58)),

$$\frac{R/d}{M_\infty^2 C_D} = \frac{0.774}{M_\infty^2 \left[C_D/(x/d) \right]^{2/3} - 1.09} \tag{61}$$

for $\gamma = 1.4$.

For $p/p_\infty \geq 1$, this equation gives values to within 4.5 percent of the second approximation (Eq. (44)).

Axisymmetric flow ($\alpha = 1$):

$$\frac{R}{d} = A \sqrt{\frac{x}{d}} \sqrt{1 - 2B \frac{x}{d}} \tag{62}$$

with

$$A = (\gamma C_D / J_0)^{1/4} / \sqrt{2}$$

and

$$B = \lambda_1 \sqrt{J_o / (\gamma C_D)} / M_\infty^2$$

hence

$$\frac{R/d}{M_\infty \sqrt{C_D}} = 0.795 \left(\frac{x/d}{M_\infty^2 \sqrt{C_D}} \right)^{1/2} \left(1 + 3.15 \frac{x/d}{M_\infty^2 \sqrt{C_D}} \right)^{1/2} \quad (63)$$

for $\gamma = 1.4$.

$$\frac{dR}{dx} = A[C/2 - B/C] \quad (64)$$

with

$$C = \sqrt{(1 - 2 Bx/d)/(x/d)}$$

$$\frac{p}{p_\infty} = \frac{\sqrt{\gamma} g^{(o)}}{8 \sqrt{J_o}} \frac{\sqrt{C_D} M_\infty^2}{x/d} \left[1 - \frac{2\lambda_1 \sqrt{J_o}}{\sqrt{\gamma}} \frac{x/d}{M_\infty^2 \sqrt{C_D}} \right]^{-1} - g^{(o)} (\lambda_1 - \psi) \quad (65)$$

$$= 0.067 \frac{M_\infty^2 \sqrt{C_D}}{x/d} \left[1 + 3.15 \frac{x/d}{M_\infty^2 \sqrt{C_D}} \right]^{-1} + 0.606 \quad (66)$$

Using the same procedure as in the case of plane flow, simpler equations for the second approximation are derived. From Eqs. (40) and (65),

$$\left(\frac{p}{p_\infty} \right)'' = \left(\frac{p}{p_\infty} \right)' \left[1 - \frac{\lambda_1 g^{(o)}}{4(p/p_\infty)'} \right]^{-1} - g^{(o)} (\lambda_1 - \psi) \quad (67)$$

$$= \left(\frac{p}{p_\infty} \right)' \left[1 + 0.211/(p/p_\infty)' \right]^{-1} + 0.606 \quad (68)$$

for $\gamma = 1.4$.

Numerical results are given in Table 7 and Fig. 1. From Fig. 1, it is evident that the simplified expression

$$\left(\frac{p}{p_\infty} \right)'' = \left(\frac{p}{p_\infty} \right)' + 0.44 \quad (69)$$

$$= 0.067 \frac{M_\infty^2 \sqrt{C_D}}{x/d} + 0.44 \quad (70)$$

gives the second approximation pressure ratio to better than one percent when $p/p_\infty > 1$, $\gamma = 1.4$.

TABLE 7
PRESSURE RATIOS AT SAME x/d
($a = 1, \gamma = 1.4, \text{FIRST AND SECOND APPROXIMATIONS}$)

$(p/p_\infty)''$	0.606	0.958	1.432	2.415	5.404	10.4
$(p/p_\infty)'$	0	0.5	1	2	5	10

For the shock shape, Eqs. (10) and (26) give

$$\frac{R}{d} = \sqrt{\frac{\gamma g^{(o)}}{16 J_o}} M_\infty \sqrt{C_D} \left[p/p_\infty + (\lambda_1 - \psi) g^{(o)} \right]^{-1/2} \tag{71}$$

$$= 0.206 M_\infty \sqrt{C_D} (p/p_\infty - 0.606)^{-1/2} \tag{72}$$

Using Eq. (69), the simplified expression is

$$\frac{R}{d} = \frac{0.795 M_\infty \sqrt{C_D}}{\left[M_\infty^2 \sqrt{C_D} / (x/d) - 2.478 \right]^{1/2}} \tag{73}$$

For $p/p_\infty > 1$, this expression gives values to better than one percent of the second approximation, Eq. (63).

SUMMARY OF BLAST ANALOGY SOLUTIONS FOR $\gamma = 1.4$

For convenience, the above-derived results for $\gamma = 1.4$ are listed as follows:

Plane Flow

First approximation:

$$\frac{R/d}{M_\infty^2 C_D} = 0.774 \frac{(x/d)^{2/3}}{M_\infty^2 C_D^{2/3}} \tag{29}$$

$$\frac{p}{p_\infty} = 0.121 M_\infty^2 \left[C_D / (x/d) \right]^{2/3} \tag{33}$$

Second approximation (simplified expressions):

$$\frac{R/d}{M_\infty^2 C_D} = \frac{0.774}{M_\infty^2 C_D^{2/3} / (x/d)^{2/3} - 1.09} \tag{61}$$

$$p/p_\infty = 0.121 \frac{M_\infty^2 C_D^{2/3}}{(x/d)^{2/3}} + 0.56 \tag{58}$$

Axisymmetric Flow

First approximation:

$$\frac{R/d}{M_\infty \sqrt{C_D}} = 0.795 \sqrt{\frac{x/d}{M_\infty^2 \sqrt{C_D}}} \quad (37)$$

$$p/p_\infty = 0.067 \frac{M_\infty^2 \sqrt{C_D}}{x/d} \quad (41)$$

Second approximation:

Exact:

$$\frac{R/d}{M_\infty \sqrt{C_D}} = 0.795 \left(\frac{x/d}{M_\infty^2 \sqrt{C_D}} \right)^{1/2} \left(1 + 3.15 \frac{x/d}{M_\infty^2 \sqrt{C_D}} \right)^{1/2} \quad (63)$$

Simplified:

$$\frac{R/d}{M_\infty \sqrt{C_D}} = 0.795 / \left[M_\infty^2 \sqrt{C_D} / (x/d) - 2.478 \right]^{1/2} \quad (73)$$

Simplified:

$$p/p_\infty = 0.067 \frac{M_\infty^2 \sqrt{C_D}}{x/d} + 0.44 \quad (70)$$

OTHER BLAST AND STEADY-FLOW SOLUTIONS

The original Taylor's (Ref. 3) solution (first approximation) of a spherical blast was extended by Lin (Ref. 6) to the case of a cylindrical blast. Lin also obtained the axisymmetric, steady-flow solution (first approximation) analogous to the cylindrical shock. At about the same time, Sakurai (Refs. 4, 5) published first and second approximation solutions of plane, cylindrical, and spherical blasts. Cheng and Pallone (Ref. 7) applied the plane wave, first approximation solution to the case of steady flow past a blunt plate, for $\gamma = 1.4$ and 1.667 . They also suggested that, in order to extend the validity of the solution to higher x/d values, the pressure should be interpreted as an increment over the free-stream pressure rather than as the absolute value.

In Table 8, the numerical values of coefficients quoted by the above-mentioned authors are compared with the values given by Sakurai (Ref. 4) and used in this paper.

TABLE 8
VALUES OF COEFFICIENTS
IN THE FIRST BLAST-WAVE APPROXIMATION ($\gamma = 1.4$)

Coefficient	J_0		$g^{(0)}$	
	0	1	0	1
Sakurai (Ref. 4)	1.696	0.877	0.455	0.424
Cheng and Pallone (Ref. 7)	1.121*		0.319	
Lin (Ref. 6)		0.858		0.4317

*Cheng et al (Ref. 8) pointed out that this value is in error.

Lees and Kubota (Refs. 9 and 10) applied Sakurai's results to obtain second approximation, steady-flow solutions analogous to the case of cylindrical blast. They derived "simplified" expressions for the case of a hemisphere-cylinder, as follows:

$$p/p_\infty = 0.0655 M_\infty^2 / (x/d) + 0.405 \quad (74)$$

$$R/d = 0.78 \sqrt{x/d} \left[1 + 1.62 (x/d) / M_\infty^2 \right] \quad (75)$$

These expressions correspond to the binomial expansion of Eqs. (66) and (63), using the modified-Newtonian drag coefficient for a hemisphere at $M_\infty = 7.7$, $\gamma = 1.4$ ($C_D = 0.914$).

Love (Ref. 11) proposed a method of calculation of bluntness-induced, inviscid, steady-flow pressures downstream of hemispherical and hemicylindrical noses. In his method, a value of the pressure at the shoulder is calculated or assumed, and the pressure decay as given by the first approximation, blast-wave analogy is taken. Love's expression in terms of free-stream pressure, p_∞ , is as follows:

$$p/p_\infty = \left[1 + (x'/d)^a \right]^{-b} + \left[1 + (x'/d)^{-a} \right]^{-b} \quad (76)$$

where

$$b = p_{sh}/p_\infty = (p_{sh}/p_o') (p_o'/p_\infty)$$

and

p_{sh} = shoulder pressure, i. e., pressure at the juncture of hemispherical (or hemicylindrical) nose to the cylindrical (or slab) afterbody;

p_o' = stagnation point pressure;

x'/d = dimensionless distance measured from the shoulder.

Theoretical calculations indicate that, at Mach numbers ≥ 5 , p_{sh}/p_o' remains approximately constant and equal to 0.045 in axisymmetric flow and to 0.117 in two-dimensional flow ($\gamma = 1.4$). The values of exponent a are taken as 1 and 2/3, respectively, in these two cases, following the blast analogy. Using these values, Love's expressions can be written as follows:

1. hemicylinder - flat plate:

$$p/p_\infty = 0.117 \left[1 + (x'/d)^{2/3} \right]^{-1} (p_o'/p_\infty) + \left[1 + (x'/d)^{-2/3} \right]^{-0.117} p_o'/p_\infty \quad (77)$$

2. hemisphere-cylinder:

$$p/p_\infty = 0.045 (1 + x'/d)^{-1} p_o'/p_\infty + \left[1 + (x'/d)^{-1} \right]^{-0.045} p_o'/p_\infty \quad (78)$$

The value of p_o'/p_∞ is a function of the free-stream Mach number.

COMPARISON OF THEORETICAL AND EXPERIMENTAL RESULTS WITH APPROXIMATE SOLUTIONS

The various blast analogy and other approximations listed above are compared here with three kinds of data:

1. results of theoretical calculations of flow about simple bodies, for ideal and real air;
2. measurements from wind and shock tunnel experiments;
3. results of correlations, in terms of blast-wave analogy parameters, of theoretically calculated flows.

As regards experimental data, only those measurements made at Reynolds numbers high enough to render the viscous effects negligible are included.

PLANE FLOW

On the basis of calculation (method of characteristics) and correlation of flow about flat plates with sonic wedge leading edges, Baradell and Bertram (Ref. 12) suggested the following expression* for pressure distribution ($\gamma = 1.4$):

$$p/p_\infty = 0.117 M_\infty^2 \left[C_D/(x/d) \right]^{2/3} + 0.732 \quad (79)$$

*This is derived from Eq. (7) of Ref. 12, with $(p - p_\infty)/p_\infty$ increased by 3 percent for $\gamma = 1.4$, as suggested by the authors.

This formula is quite similar (see Figs. 2 and 8) to the simplified second approximation, Eq. (58), and gives slightly larger values of p/p_∞ at small values of $M_\infty^2 [C_D/(x/d)]^{2/3}$, and vice versa. The formula was derived on the basis of two calculations, both for $C_D \approx 1.4$ and for $M_\infty = 9.5$ and $M_\infty = 20$.

Pressure distribution measured on a flat plate with a semi-cylindrical leading edge is compared with various approximations in Fig. 2. The experimental data extend to $x/d = 10.5$. Within this range, all blast-wave approximations calculated for the experimental pressure drag coefficient of $C_D = 1.27$ show reasonable agreement with experimental values at $x/d > 1.5$. The correlation formula of Ref. 12, Eq. (79), gives values very slightly larger than the second approximation. Although Love's method predicts correctly the shoulder pressure, it is least accurate at intermediate x/d values.

Also in Fig. 2 a curve is included showing second blast approximation computed for $M_\infty = 8$ and $C_D = 1$ instead of the experimental value of 1.27. The agreement with measurements, although presumably fortuitous, is excellent.

In Figs. 3, 4, and 5, approximations to the pressure distribution and shock shape in plane flow are compared with the theoretical results obtained from calculations using the method of characteristics.

The superiority of the second approximation to the pressure distribution at large x/d values for a configuration having a high nose drag is very evident at $M_\infty = 6.86$ from Figs. 3a and b. The difference between the theoretical and approximate pressure ratios at any given x/d is substantially constant, and the approximate pressure ratio is within 20 percent of the theoretical value even at $x/d = 150$. A still better agreement, to within 10 percent, is obtained with the correlation formula, Eq. (79).

In Fig. 4, the corresponding theoretical and approximate shock shapes are shown, and the superiority of the second blast approximation is again evident. This is also the case at $M_\infty = 12.3$, Fig. 5; the second blast approximation gives values to better than 10 percent of the experimental observations.

Results of flow field computations for real air obtained by the General Electric Company (Refs. 16 and 17) are shown in Figs. 6 and 7. The blast analogy does not predict the pressure distribution accurately for $\gamma = 1.4$; a much better agreement was obtained for $\gamma = 1.3$, Fig. 6. The shock slope, as in the previous cases, is well approximated by blast analogy for $\gamma = 1.4$. In Fig. 7, the shock Mach numbers, M_R , computed

from Eqs. (24) and (31), are shown and indicate that real gas effects might affect a significant portion of the flow field.

The data shown in Figs. 2 through 7 are plotted in terms of blast analogy correlation parameters in Figs. 8 and 9.

All theoretical results for pressure distribution, for $\gamma = 1.4$, Fig. 8, correlate well with Eq. (79). The experimental data fall below the correlation of theoretical data. The inadequacy of the first blast approximation is again clearly seen.

Figure 8 indicates that, although blast analogy provides suitable correlation parameters, the range of its applicability, in terms of x/d , depends on values of C_D and M_∞ . For example, with $C_D = 1.4$, the prediction of pressure ratio becomes poor at $M_\infty = 9.5$ when $(x/d)^{2/3} / (M_\infty^2 C_D^{2/3}) < 0.025 (x/d < 5)$, whereas at $M_\infty = 20$, it is good down to about $0.004 (x/d \approx 3)$.

Comparison of results of theoretical calculations for ideal ($\gamma = 1.4$) and real air, for Mach numbers of 20 and 15, respectively, presumably indicates the already noted significant real gas effects.

In Fig. 9, the available data on shock shape are plotted in terms of correlation parameters of the second approximation blast analogy, together with the theoretical curve. The two sets of data correlate poorly. The experimental results agree well with the theoretical curve at large values of x/d .

From the above review of the limited data available for plane flow, it is apparent that blast analogy provides useful correlation of pressure distributions and shock shapes. Pressures are well predicted by the second blast analogy approximation for ideal gas, $\gamma = 1.4$; results of a calculation for real air at $M_\infty = 15$ indicate pressures lower by 20 percent or more. Shock slopes are well predicted by the second blast analogy approximation, whereas the shock location shows an outward displacement relative to blast analogy prediction.

AXISYMMETRIC FLOW

Experimental Data

The available experimental pressure distributions on hemisphere cylinders at Mach numbers 6, 7.7, and 8 are compared with blast analogy and Love's (Ref. 11) predictions in Figs. 10a, b, and c. In all cases at $x/d > 3$, the second blast analogy approximation gives values much more

realistic than the first approximation and predicts the pressure distribution accurately at $M_\infty = 7.7$ and 8 . Love's method gives the pressure level accurately at $M_\infty = 6$, $x/d > 4$, but is less accurate than the second blast analogy approximation at $M_\infty = 8$.

Pressure distribution on a flat-nose cylinder at $M_\infty = 8$ is shown in Fig. 11. Within the limited range of measurements ($x/d \leq 5.5$), both first and second blast analogy approximations give an accurate estimate of the pressure on this high drag ($C_D = 1.64$) body.

The experimental and blast analogy data on shock shapes are given in Fig. 12 for two of the cases already considered. As in the plane flow, the second approximation predicts closely the shock slope.

Lees and Kubota (Refs. 9 and 10) measured the radial pitot pressure distribution around the hemisphere-cylinder model at $M_\infty = 7.7$ at station $x/d = 3$ and computed the corresponding distributions of other quantities. These results for p/p_∞ and ρ/ρ_∞ are shown in Figs. 13 and 14 together with the corresponding blast analogy predictions. The latter were obtained from Eqs. (3) and (8) with $y = 1/M_R^2$ computed from Eqs. (24), (39), and (64).

The experimental pressure distribution, Fig. 13, shows an excellent agreement with the second blast approximation over the whole range of r/R . At the shock, the observed pressure ratio is 5, whereas the one computed from the second approximation blast analogy is 5.13.

The density distributions are compared in Fig. 14. As mentioned previously, the second blast approximation gives the exact density ratio across the shock only at high shock Mach numbers. In this case, $M_R \approx 2.13$, and hence a rather poor agreement is obtained with the experimental distribution. However, the exact ρ/ρ_∞ ratio for $M_R = 2.13$ equals 2.85 and agrees well with the experimental value (Fig. 14).

Nevertheless, in view of the good agreement in the pressure distributions in Fig. 13, it can be considered that the flow field closely approximates the second approximation blast solution. It is then possible to write, for the density distribution, an expression which is exact at the shock and to determine a function $\chi'(\kappa)$ from the experimental data. Thus, by analogy to Eq. (17),

$$\rho/\rho_\infty = h^{(o)}(\kappa) / [1 + \chi'(\kappa) y] \quad (80)$$

where $h^{(o)}(\kappa)$ is given by the first blast approximation and χ' is computed to fit the experimental density distribution of Fig. 14. The values of χ' are

given in Table 9 and were computed assuming $M_R = 2.11$, which corresponds to the experimental pressure and density ratios across the shock. At higher shock Mach numbers, Eqs. (80) and (8) are in close agreement.

TABLE 9
VALUES OF χ' ($M_R = 2.11$)

r/R	1	0.95	0.9	0.8	0.7	0.6	0.5	0.4
χ'	5	1.75	0.1	-1.45	-2.45	-3.2	-3.8	-4.25

Theoretical Data for $\gamma = 1.4$

Van Hise (Ref. 14) published results of calculations of flow fields about cylinders with a variety of nose shapes, ranging in drag coefficients from 0.037 to 1.37. He used the method of characteristics to obtain surface pressure distributions and shock shapes in the flow of perfect gas with $\gamma = 1.4$ and $5/3$. In most cases, the nose drag coefficient was computed from the surface pressure distribution*.

Pressure distributions obtained by Van Hise are shown in Figs. 15a, b, and c in terms of the blast analogy correlation parameter, $(x/d)/(M_\infty^2 \sqrt{C_D})$. Also in these figures is included the curve given by

$$p/p_\infty = 0.06 M_\infty^2 \sqrt{C_D} / (x/d) + 0.55 \quad (81)$$

which was derived by Van Hise as representing the best correlation of all of his results for air ($\gamma = 1.4$). Equation (81) above closely approximates the theoretical blast analogy expression, Eq. (70) (see Fig. 24).

It is immediately apparent from Van Hise's results that the blast analogy provides an excellent means of correlating the pressure data over a wide range of drag and Mach numbers. It is also apparent that the range of validity of an expression such as Eq. (81) is limited, in each case, to a certain range of the correlation parameter. Based on Figs. 15b and c, for bodies with $C_D \approx 1$ and at $M_\infty \geq 10$, Eq. (81) predicts accurately the pressure distribution at $x/d \geq 2$. The lower limits (in terms of free-stream Mach number and correlation parameter, $(x/d)/(M_\infty^2 \sqrt{C_D})$), of validity of blast analogy correlation are indicated in more detail in Fig. 16 for different drag coefficients, based on agreement in the pressure ratio to

*Values of C_D assumed rather than computed are given in parentheses in Figs. 15 and 18. Only the results for $\gamma = 1.4$ are here considered.

within 10 percent. The theoretical data of Van Hise is self-consistent, whereas the experimental data indicate higher limits of the correlation parameter for given C_D and M_∞ . Similar data are presented in Fig. 17 for $M_\infty = 40$ and a wide range of C_D .

As regards the upper limits of the validity of the correlation parameter, the available data are not extensive enough to allow their determination. In the majority of cases considered, good correlation was obtained for

$$(x/d)/(M_\infty^2 \sqrt{C_D}) \leq 0.1 \quad \text{or} \quad p/p_\infty \geq 1.$$

In a number of cases in which calculations have been made into the region of $p/p_\infty < 1$ (Figs. 15b and c, at $M_\infty = 10$ and 6.9), significant deviations from Eq. (81) are evident at $p/p_\infty < 1$, but the data appear nevertheless to be well correlated by the blast analogy parameter. This, however, is not generally true, as indicated in Fig. 15c by the pressure distribution at $M_\infty = 20$, $C_D = 1.37$, which deviates from other data already at $p/p_\infty \approx 1.3$ and at smaller p/p_∞ values. As pointed out by Van Hise*, the results at $p/p_\infty < 1.3$ may be subject to significant cumulative errors.

The corresponding available results for the shock shape are shown in Fig. 18 in terms of the blast analogy correlation parameters. The correlation is good considering the wide range of drag and Mach numbers and falls slightly above the predicted $(R/d)/(M_\infty \sqrt{C_D})$ values.

Theoretical Data for Real Air in Equilibrium

Two sets of theoretical calculations of flow fields around a hemisphere-cylinder in real air in equilibrium are available for comparison with blast analogy predictions and correlations. Feldman (Ref. 15) published results of characteristic calculations for $M_\infty \approx 18$ (17,500 ft/sec) at 60,000-ft altitude, and the General Electric Company (GE) (Ref. 16) computed similar data for $M_\infty \approx 15$ to 19 at 100,000 to 200,000-ft altitude, using the flow field method of Gravalos (Ref. 17). Details of the conditions for the GE calculations are given in Table 10 on the following page.

*Private communication.

TABLE 10

**GE CALCULATIONS OF FLOW FIELDS AROUND A HEMISPHERE-CYLINDER
IN REAL AIR IN EQUILIBRIUM**

M_∞	u_∞ , ft/sec	C_D	h_∞ , ft	P_∞ , lb/ft ²	$\rho_\infty \times 10^6$, slug/ft ³	T_∞ , °R
(15)	14,532	0.897	100,000	23.085	32.114	418.79
18.24	20,000	0.914	150,000	3.0597	3.5642	500.11
18.1	20,000	0.91	175,000	1.2334	1.4123	508.79
19.25	20,000	0.908	200,000	0.47151	0.6118	449.0

Shock shapes and pressure distributions obtained by Feldman (Ref. 15) and from blast analogy predictions (for $\gamma = 1.4$, $C_D = 1$) are shown in Fig. 19. As for the other data, the shock shape agrees well with the second blast approximation, except for a substantially constant, outward displacement of about $(1/4)d$.

The pressure distribution, as given by the second blast analogy approximation, shows a fair agreement with theory at large x/d values.

The GE data are shown in Figs. 20 through 23 for $M_\infty = 15$ and 18.1. The shock shapes and, more particularly, the shock slopes are well predicted. At $M_\infty = 15$ (Fig. 21), the pressure distribution agrees well with all blast analogy approximations for $\gamma = 1.4$ at $x/d \geq 3$. At $M_\infty = 18.1$ (Fig. 23), the agreement is somewhat poorer at $x/d < 4$, probably indicating a stronger influence of real gas effects. However, compared to the results obtained by Feldman (Fig. 19b), a much better agreement with the blast analogy is evident at small x/d values.

For comparison, results for $\gamma = 1.3$ are also included in Figs. 21 and 23. It is apparent that a change in γ does not account for the differences between the calculated and predicted pressure distributions.

The real air data are shown in terms of the blast analogy parameters in Figs. 24, 25, and 26, together with the experimental results and curves representing blast analogy solutions and other correlations.

At lower values of the pressure ratio (Fig. 24) the real air results show a fair agreement with the other data. At pressure ratios above about 4, the real air results for $M_\infty = 18$ to 20 show a marked divergence from the blast analogy and the correlation of theoretical data for $\gamma = 1.4$. Also, unlike the theoretical computations for cone-cylinders and cone-sphere-cylinders (Fig. 15), the pressure distributions at $M_\infty = 18$ to 20 in real air correlate well from the shoulder ($x/d = 0.5$) downstream.

These trends are more clearly shown in Fig. 25. It is evident that, whereas the $M_\infty = 15$ results in real air are in fair agreement with the $\gamma = 1.4$ characteristic and blast analogy correlations, the $M_\infty = 18$ to 20 data correlation shows a large change in slope, which cannot result from a simple change in the value of γ .

Shock shapes for real air are compared in Fig. 26 with the theoretical blast analogy approximations ($\gamma = 1.4$) and experimental data ($\gamma = 1.4$). At larger values of x/d [$(x/d)/(M_\infty^2 \sqrt{C_D}) > 0.01$], the real air results closely approach the second approximation blast analogy prediction for $\gamma = 1.4$. As indicated in Fig. 26, the agreement would be worse for $\gamma < 1.4$. The only available experimental data fall above the blast analogy prediction, as already indicated (Fig. 12b). At small x/d values, smaller R/d values and smaller shock slope than predicted are found. This trend for real air coincides with results of the characteristic computations for $\gamma = 1.4$ (Fig. 18).

The radial pressure and density distributions computed by GE (Ref. 16) at $M_\infty = 18.1$ are compared with blast analogy predictions for $\gamma = 1.4$ in Figs. 27 and 28.

As already observed, the pressure distribution (Fig. 27) is well predicted at larger x/d values; in all cases, the first approximation agrees better with the theoretical results than the second approximation. This would be expected in view of the similarly better agreement in shock slope (Fig. 22).

The density distribution (Fig. 28) is in all cases well predicted at $r/R > 0.8$. At the shock, the second approximation gives more accurate values, and still closer ones are obtained using Eq. (80). At $r/R < 0.8$, the predicted density is much smaller than the computed one.

CONCLUDING REMARKS

Comparison of blast analogy (second approximation) solutions with correlations based on characteristic calculations for plane and axisymmetric flows shows good agreement between the two and thus provides a confirmation of the validity of the blast analogy method. The range of applicability of the blast analogy solutions depends on the values of parameters (e. g., M_∞ and C_D) which define the flow field; however, in general, at high Mach numbers and for ideal gas ($\gamma = 1.4$), the blast analogy predicts rather accurately the pressure distributions and shock shapes from

2 to 3 diameters downstream of the nose to downstream stations at which the pressure has decayed to the free-stream value. Although the shock is displaced from the body more than predicted by blast analogy, the latter gives accurate values of shock slope.

As regards application of the blast analogy method to flow of real air, the available data are limited to semi-cylinder-plate and hemisphere-cylinder configurations at $M_\infty \approx 15$ to 19, at altitudes from 60,000 to 200,000 ft. Comparisons indicate that blast analogy for $\gamma = 1.4$ predicts accurately the shock slope; the pressure distribution is predicted about as well as for experimental measurements made at lower Mach numbers (6 to 8) and temperatures.

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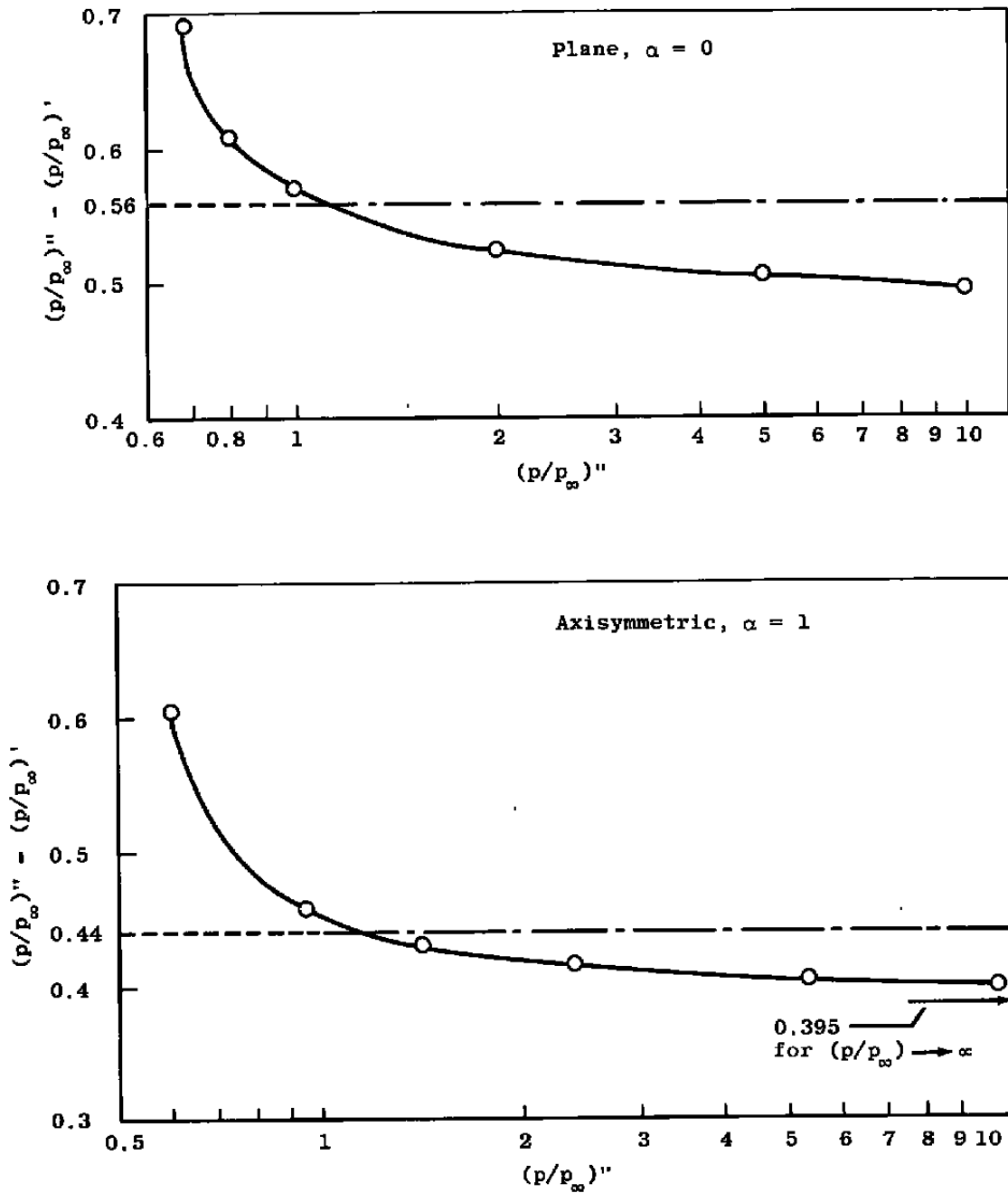


Fig. 1 Comparison of Pressure Ratios as Given by the First and Second Blast Analogy Approximations at a Given x/d Station ($\gamma = 1.4$)

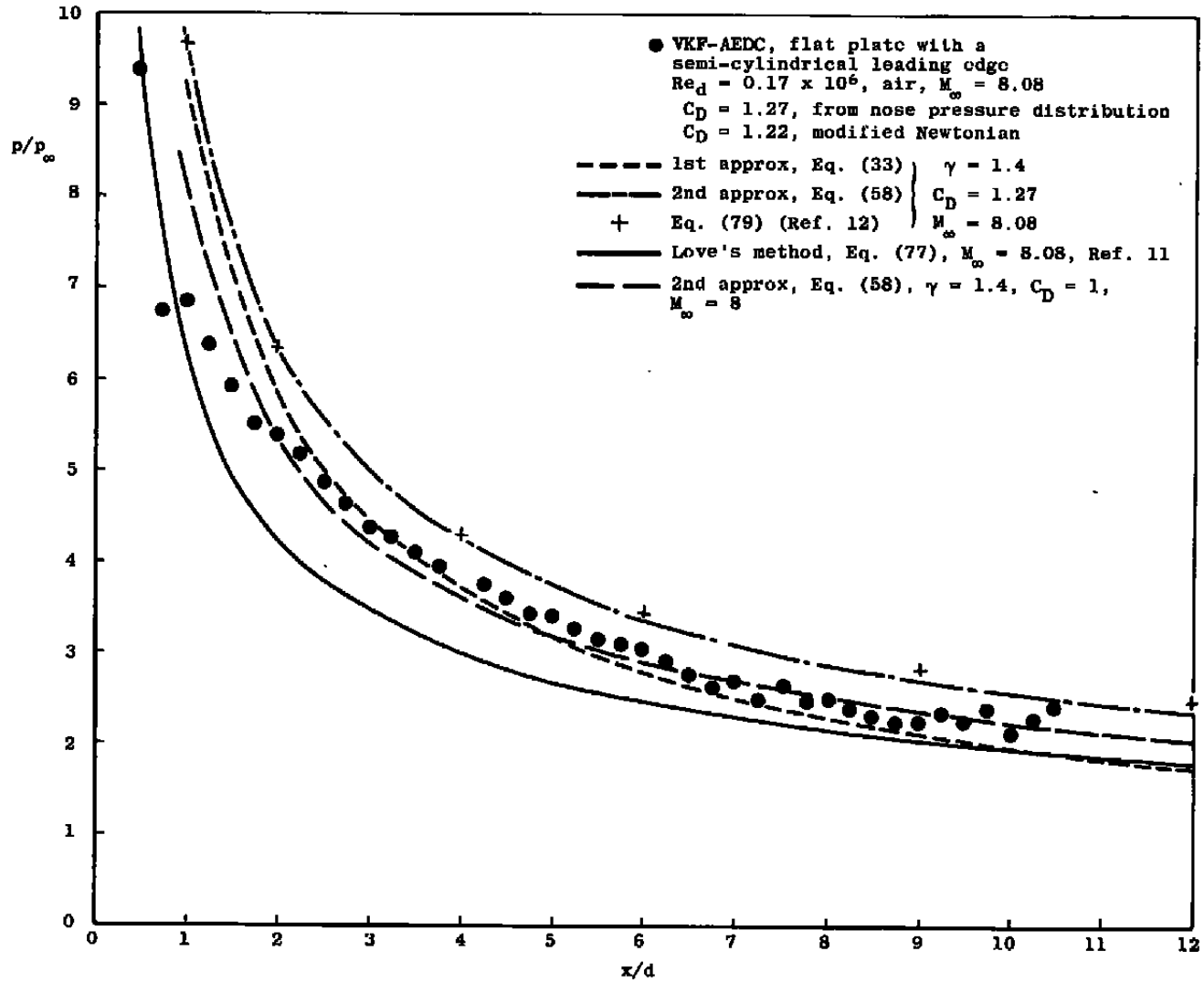
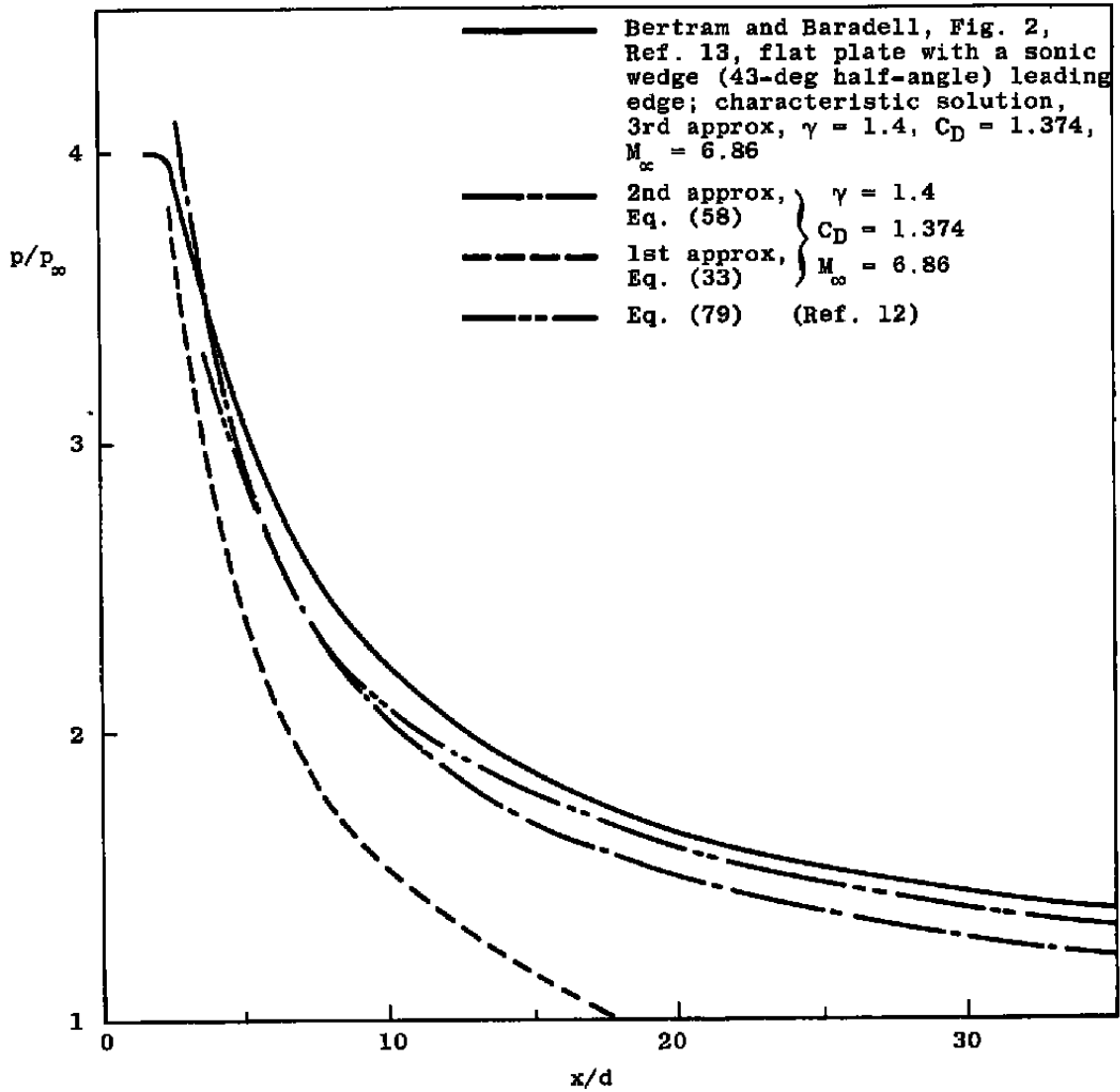


Fig. 2 Experimental and Predicted Pressure Distribution in Plane Flow, $M_\infty = 8$



a. $x/d < 35$

Fig. 3 Pressure Distribution Calculated by the Method of Characteristics and Blast Analogy in Plane Flow, $M_\infty = 6.86$

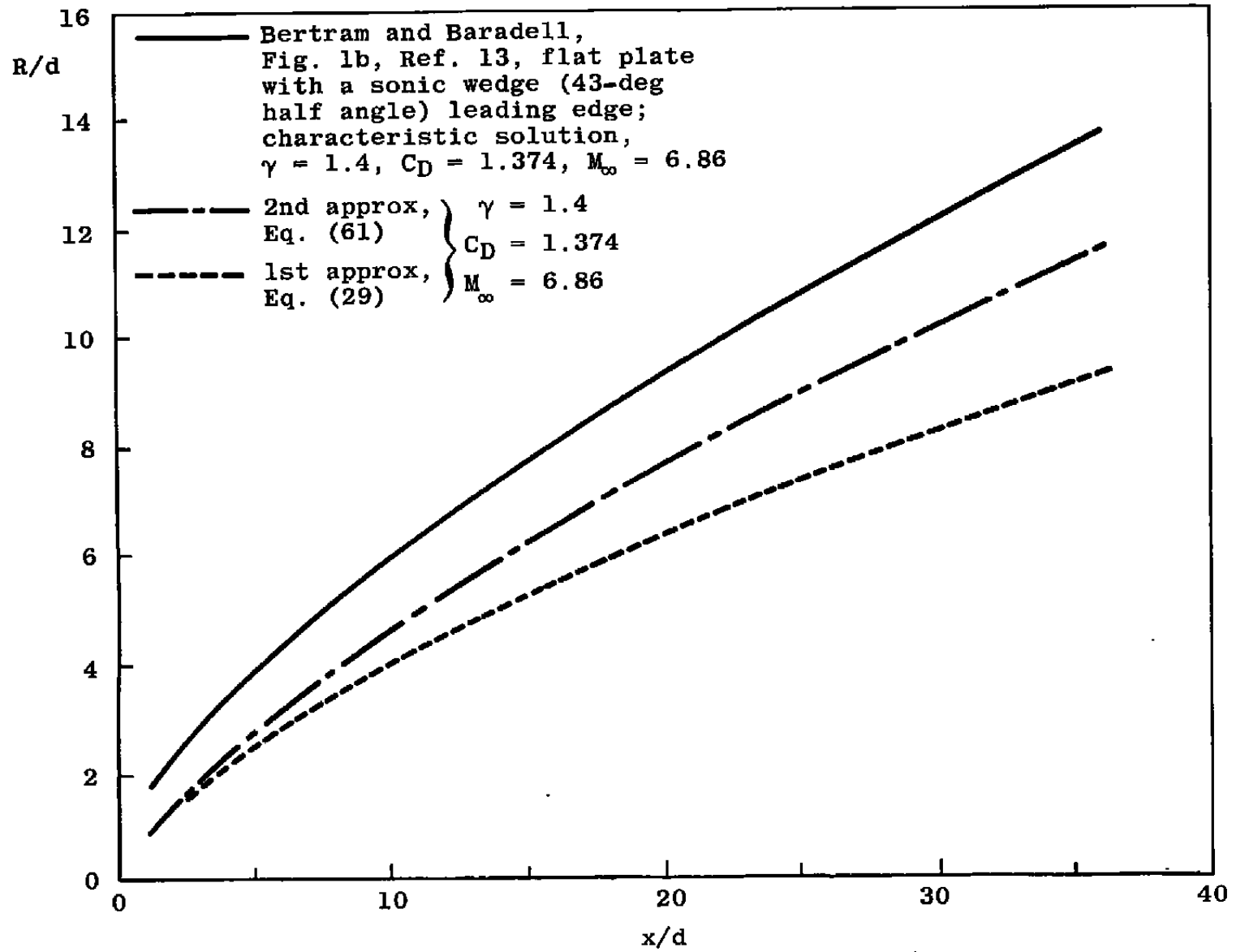
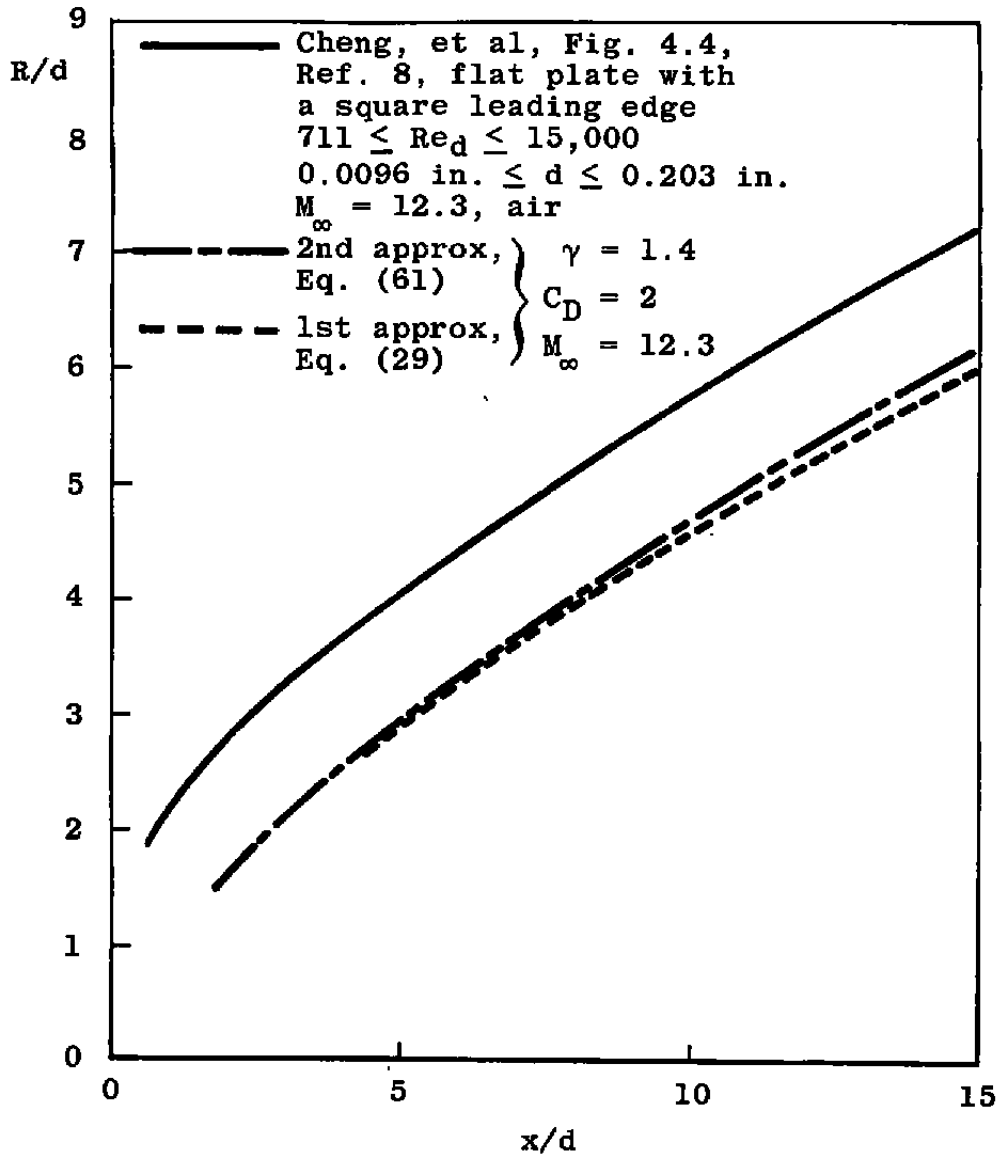
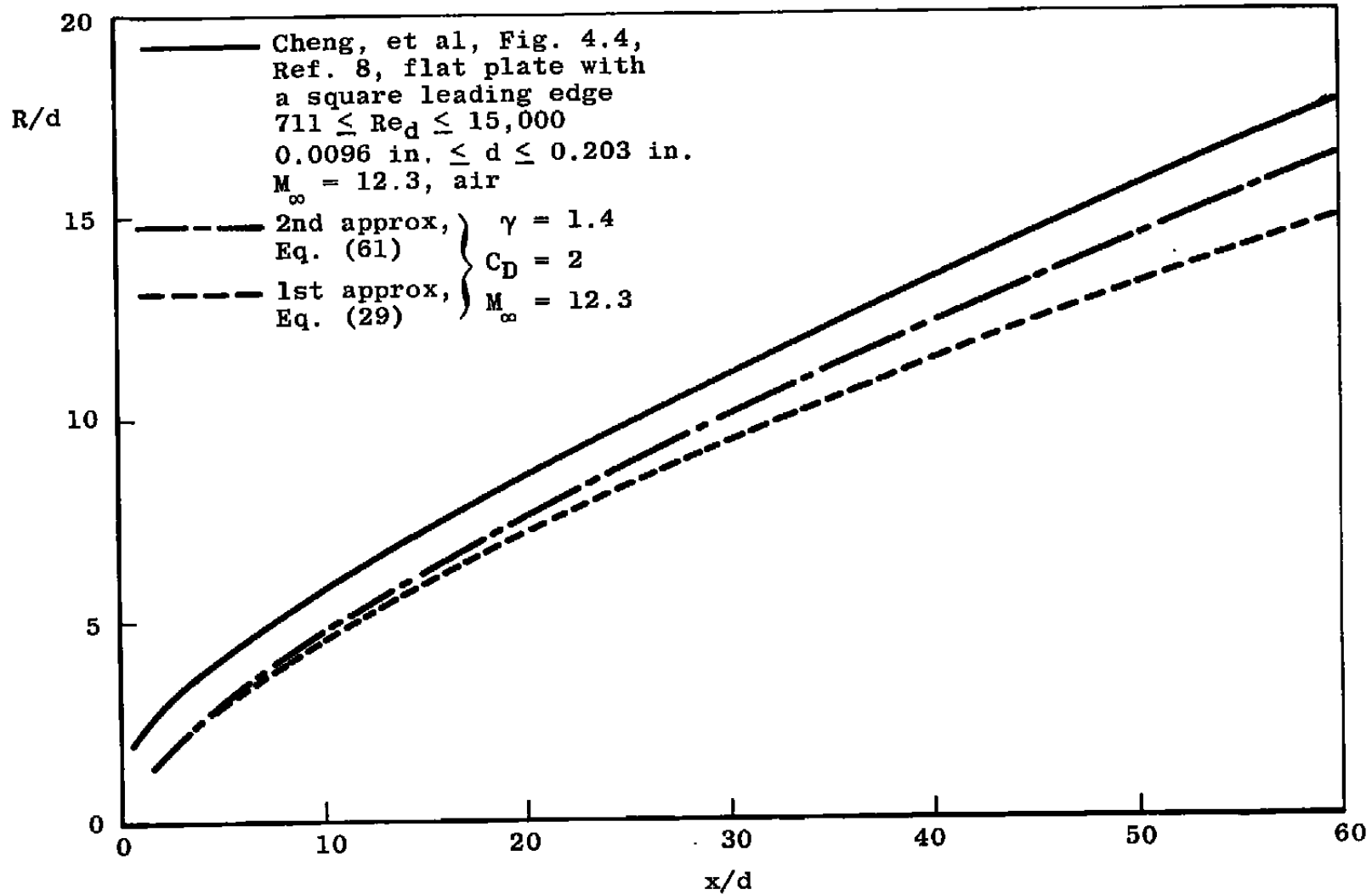


Fig. 4 Shock Shapes Calculated by the Method of Characteristics and Blast Analogy in Plane Flow, $M_\infty = 6.86$



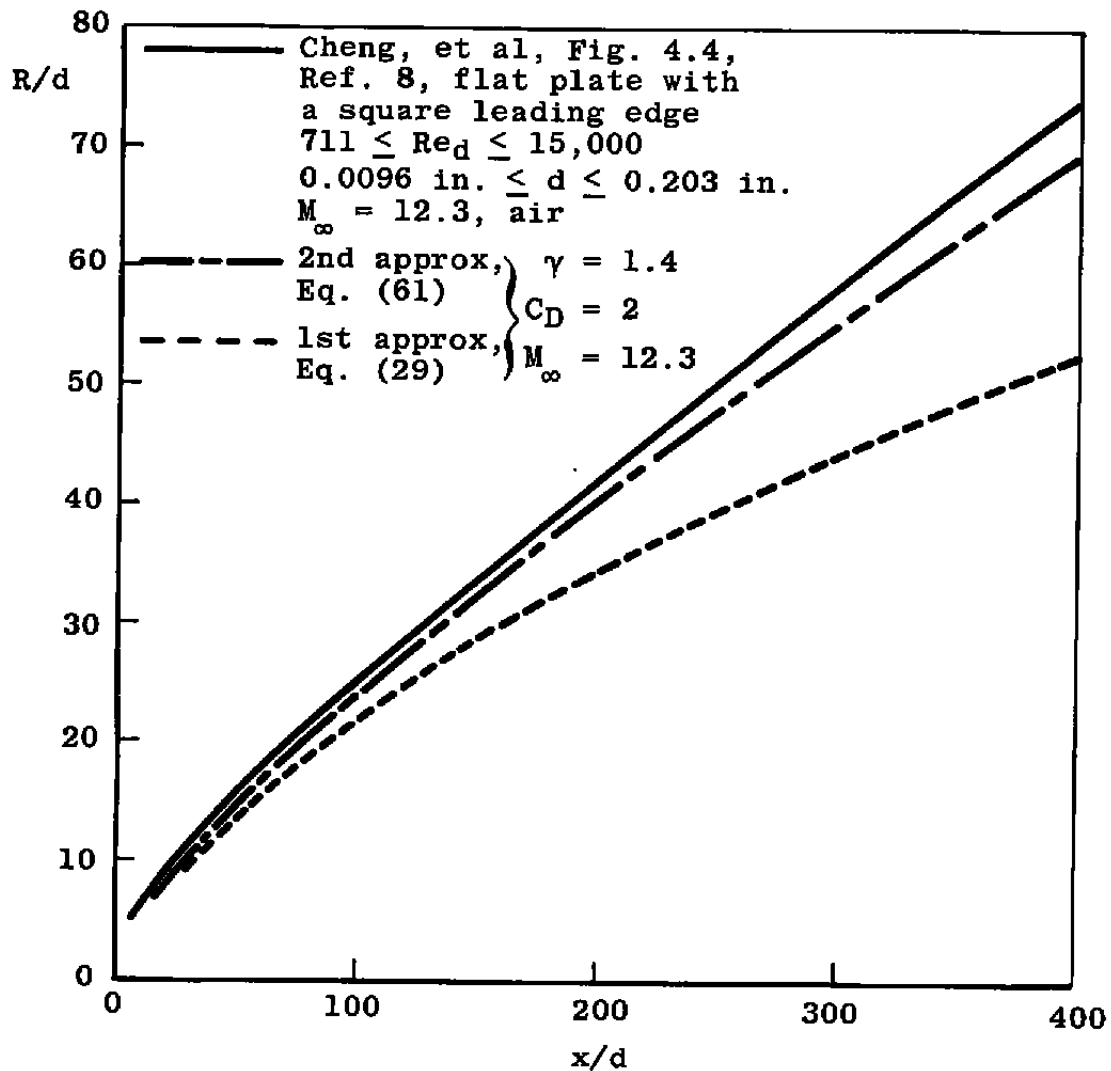
a. $x/d < 15$

Fig. 5 Experimental and Predicted Shock Shapes
in Plane Flow, $M_\infty = 12.3$



b. $x/d < 60$

Fig. 5 Continued



c. $x/d < 400$

Fig. 5 Concluded

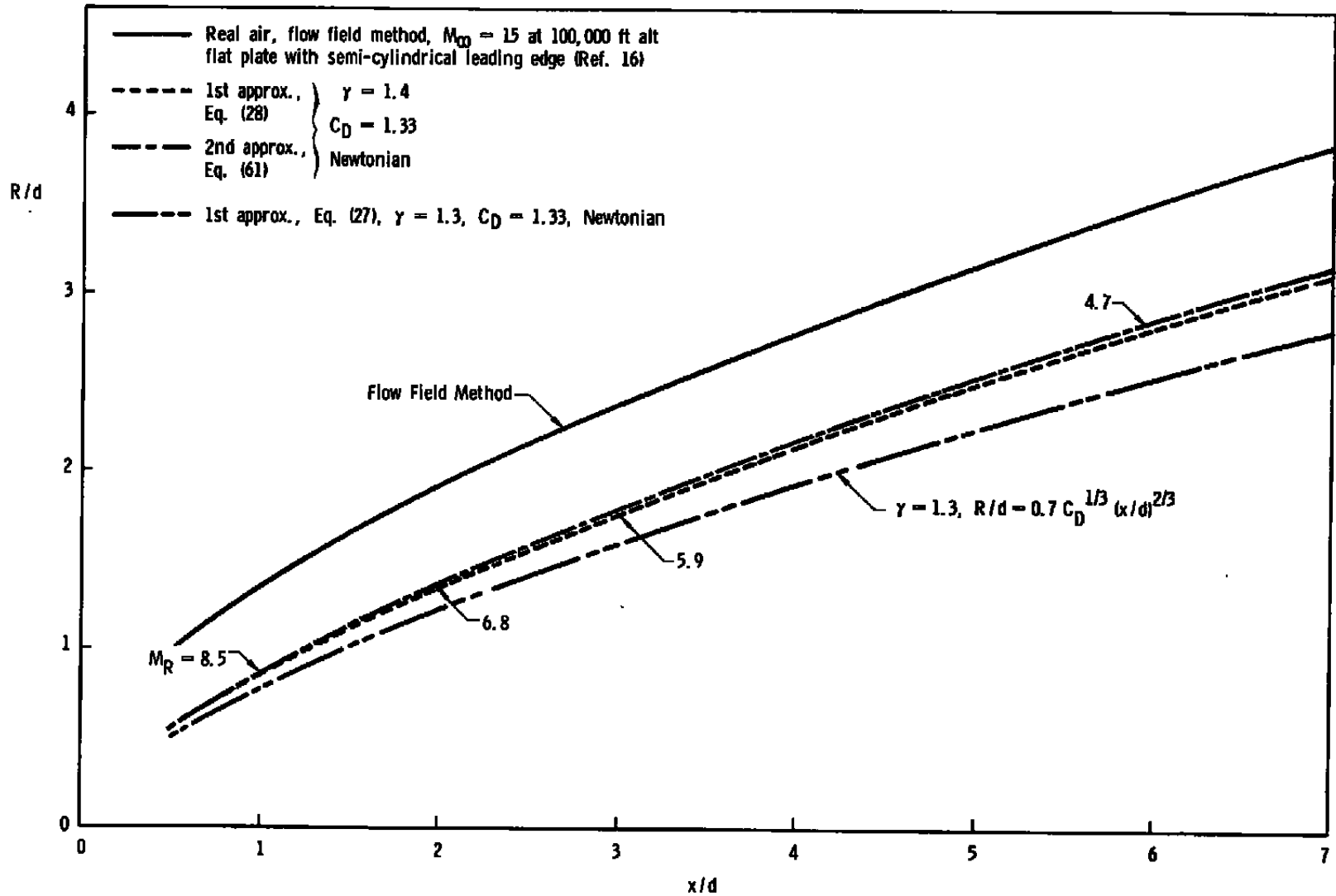


Fig. 7 Shock Shapes in Plane Flow Calculated by the Flow Field Method for Real Air and by Blast Analogy, $M_\infty = 15$

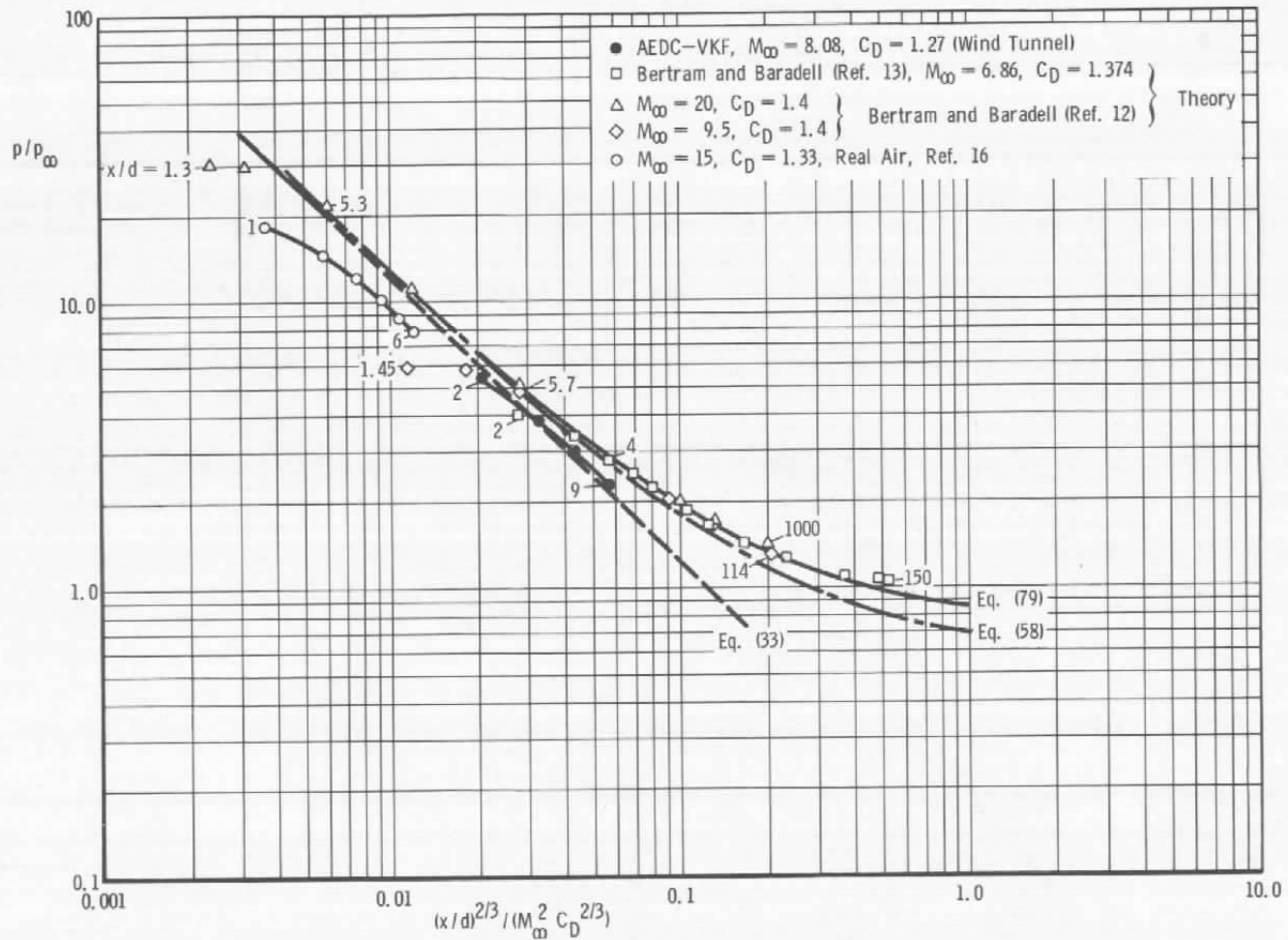


Fig. 8 Pressure Distribution in Plane Flow Correlated by Blast Analogy Parameters

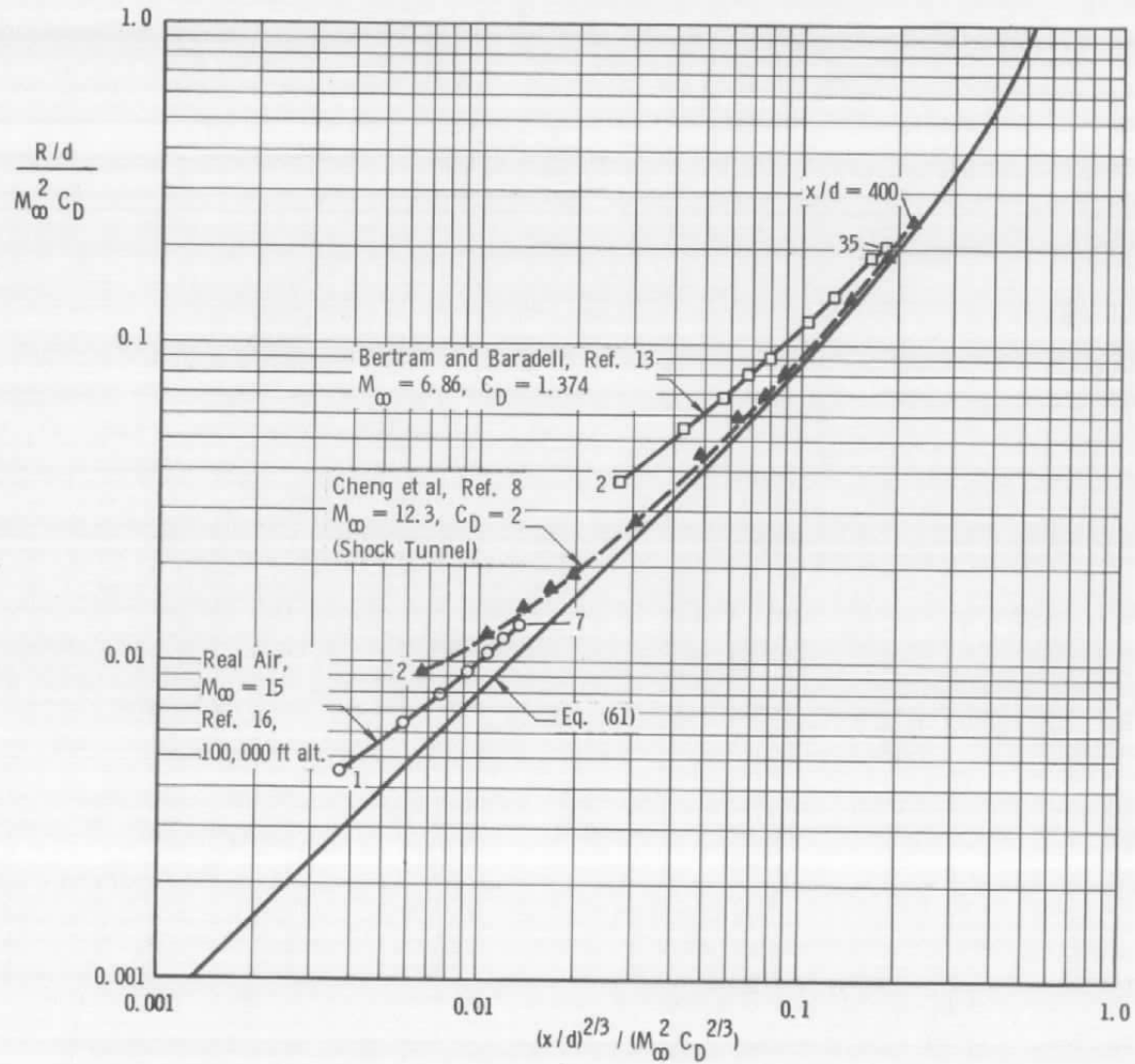


Fig. 9 Shock Shape in Plane Flow Correlated by Blast Analogy Parameters

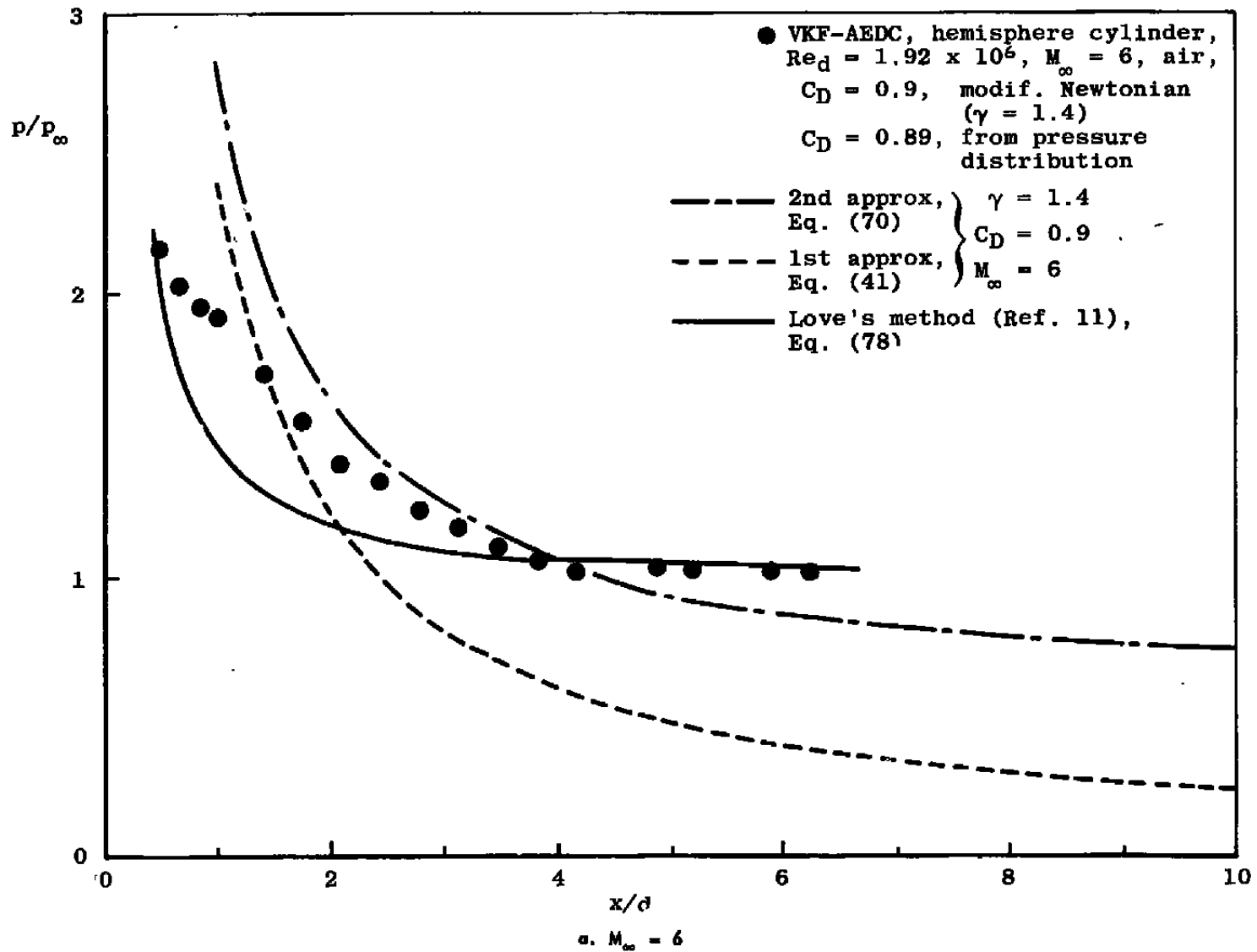
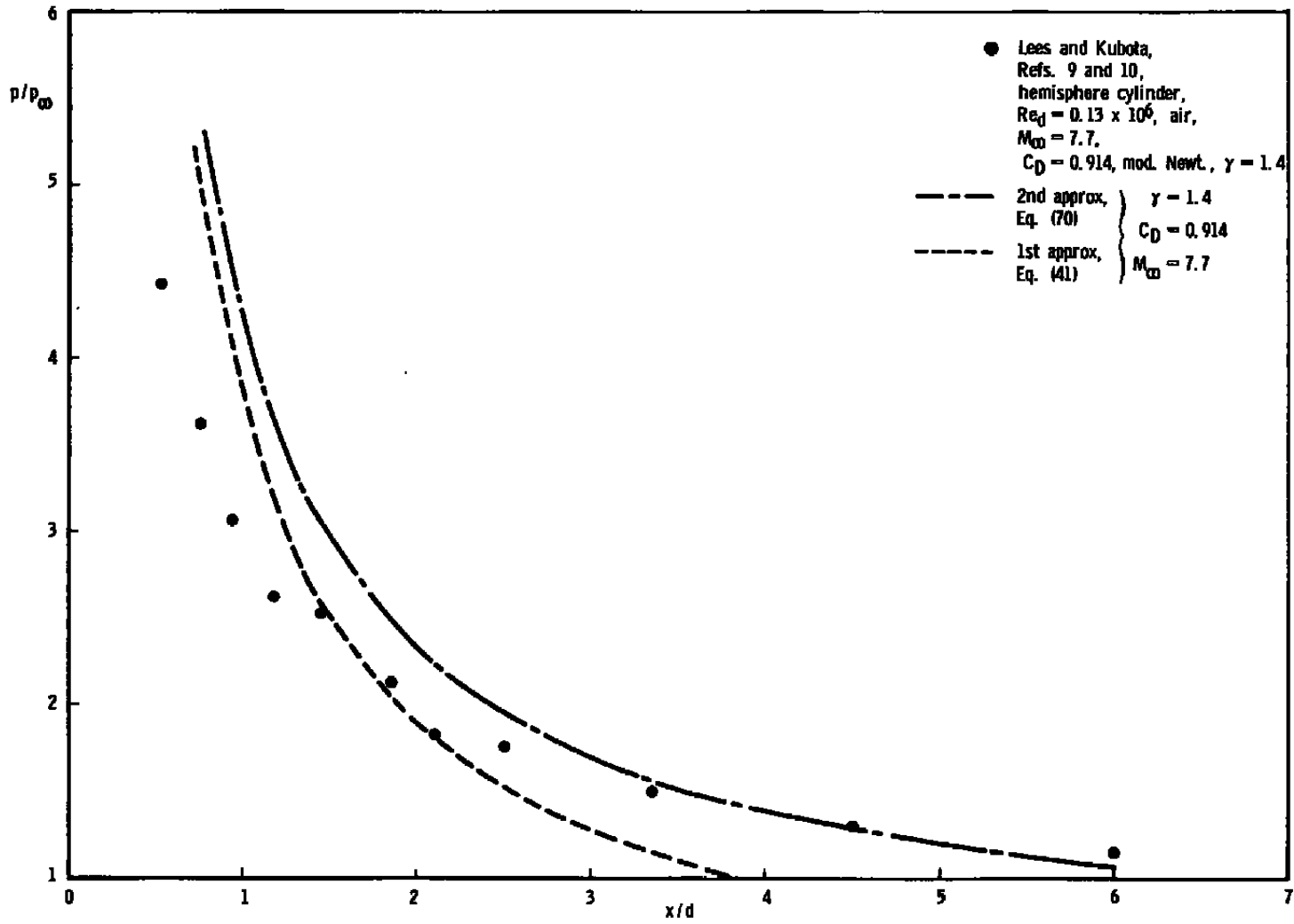


Fig. 10 Experimental and Predicted Pressure Distribution on a Hemisphere Cylinder



b. $M_\infty = 7.7$

Fig. 10 Continued

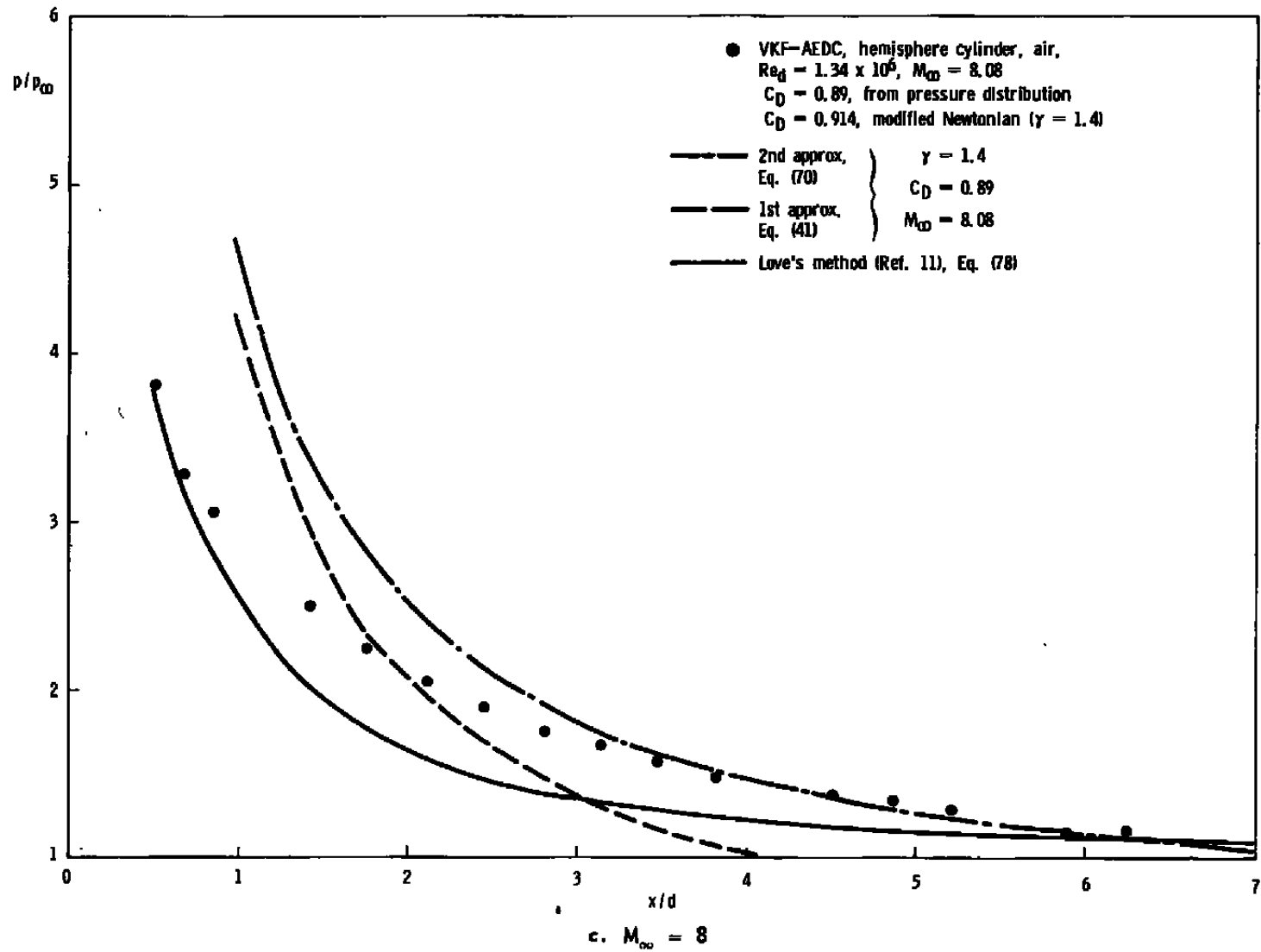


Fig. 10 Concluded

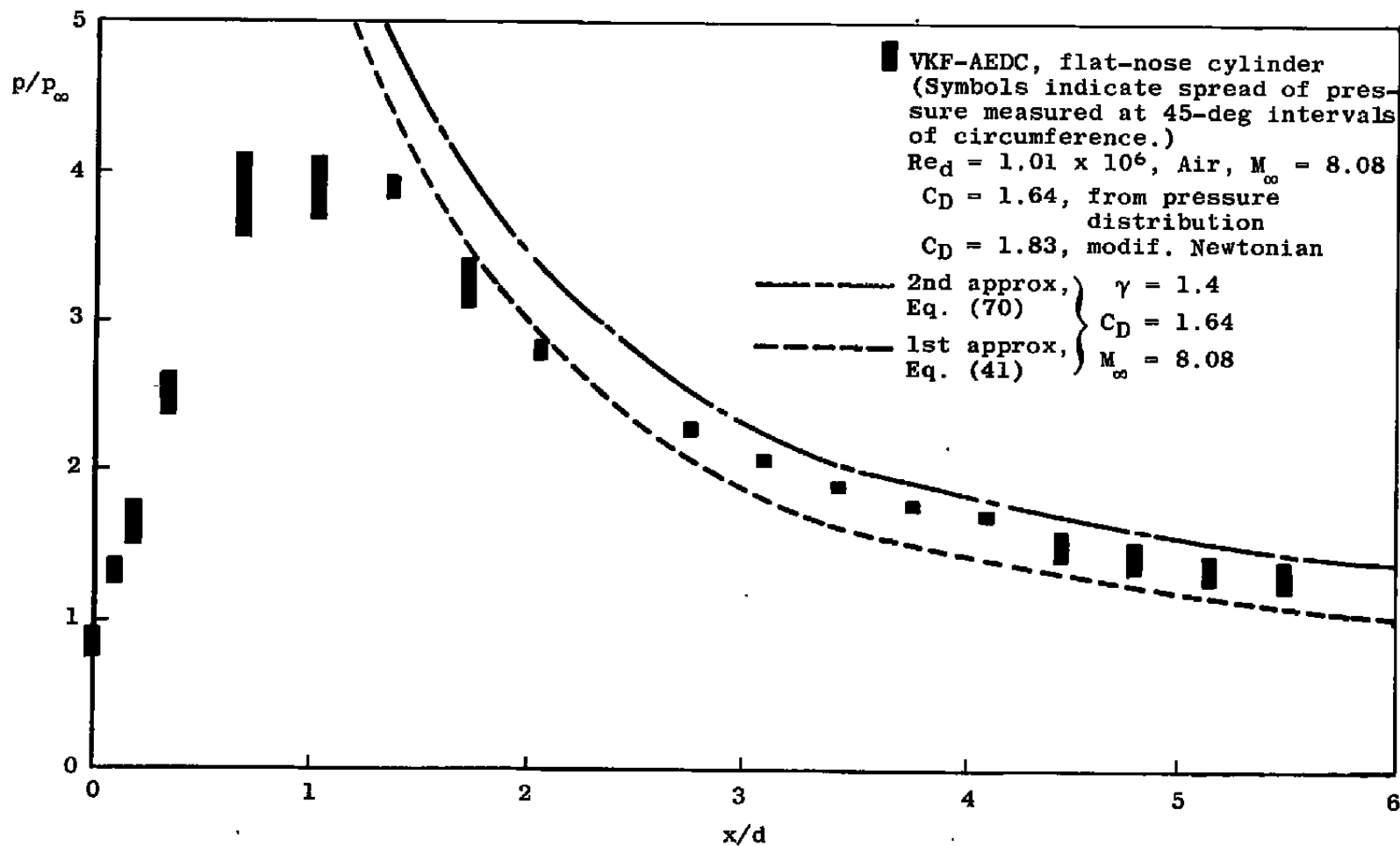


Fig. 11 Experimental and Predicted Pressure Distributions on a Flat-Nosed Cylinder, $M_\infty = 8$

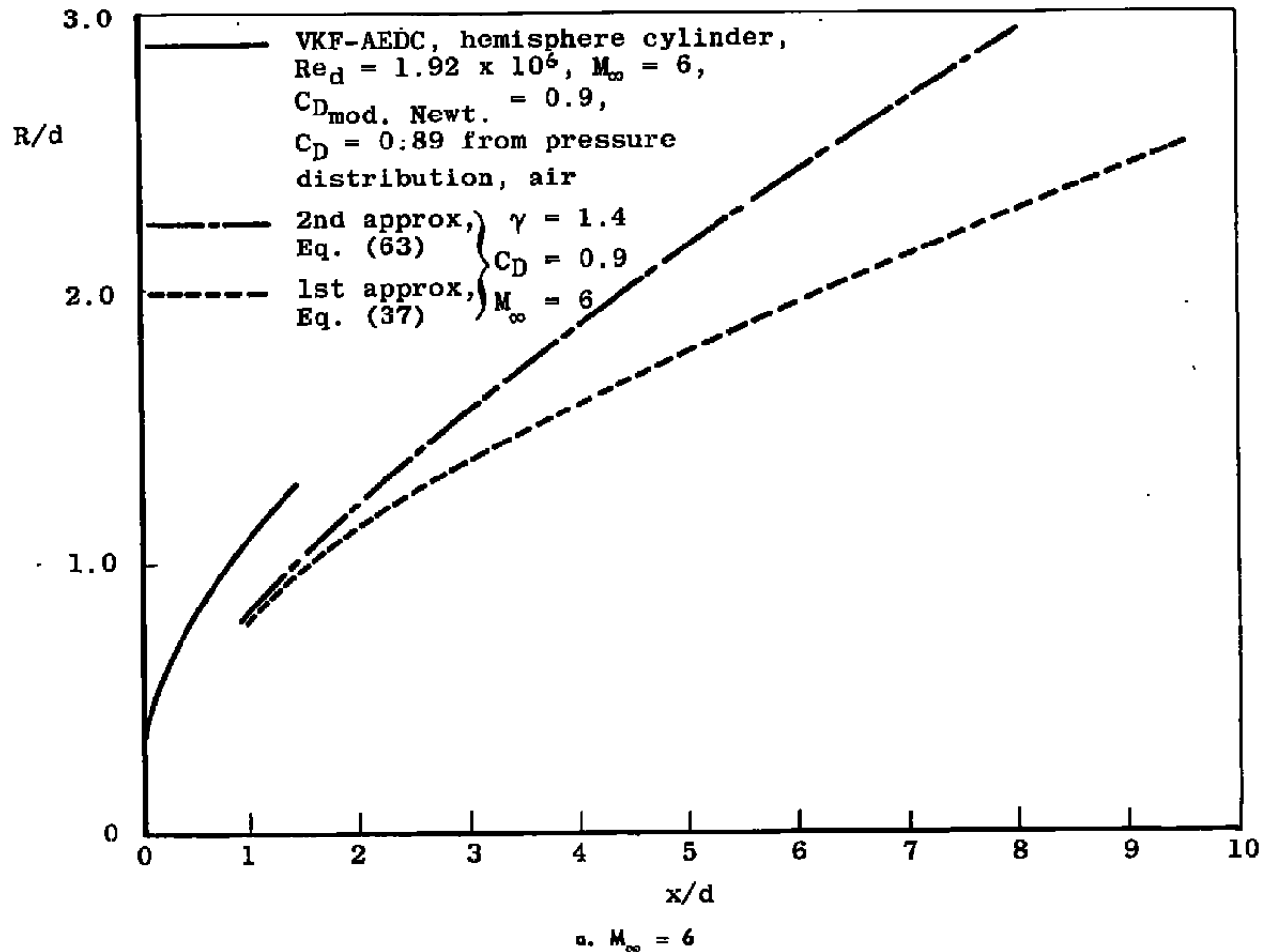
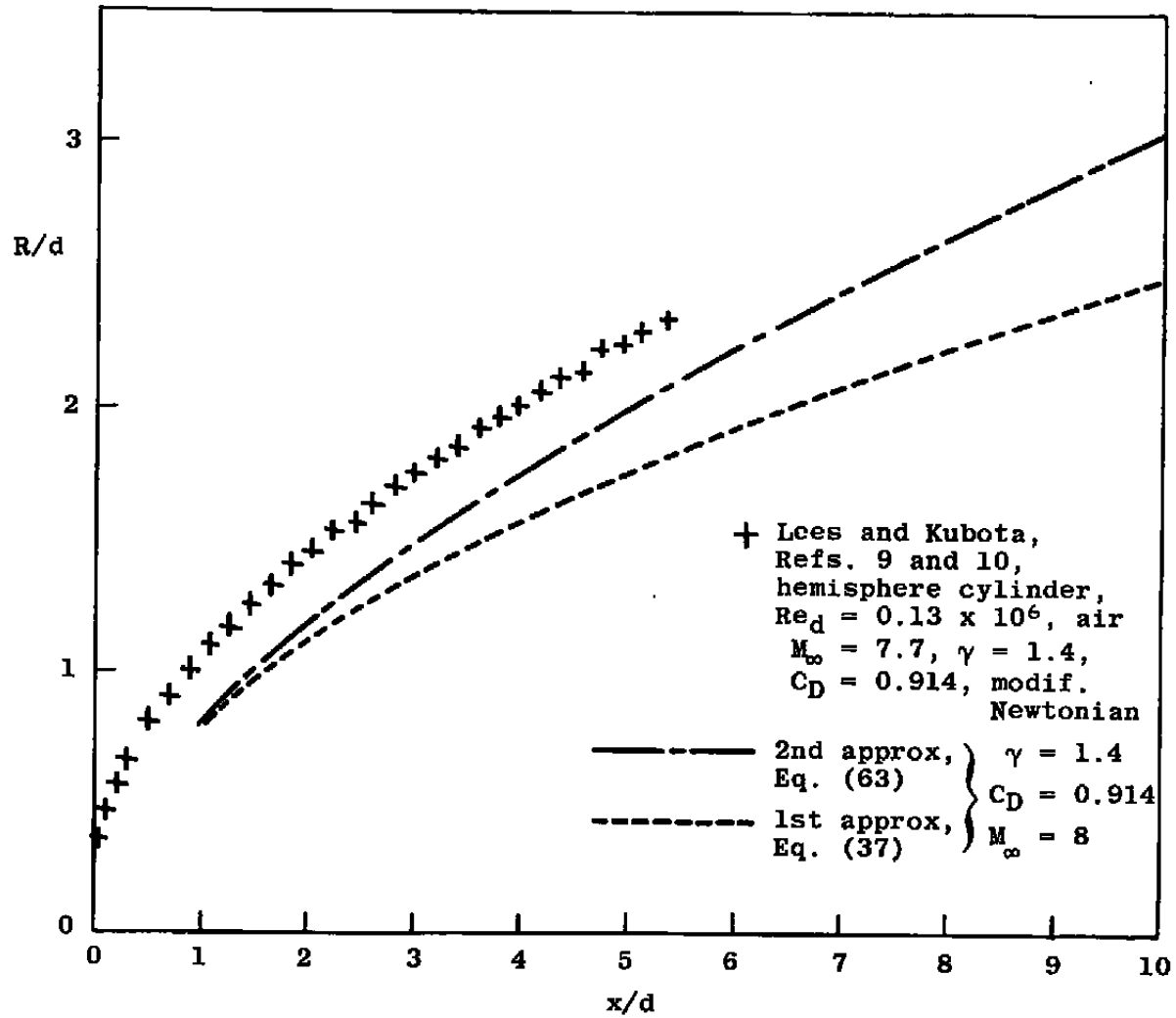


Fig. 12 Experimental and Predicted Shock Shapes around a Hemisphere Cylinder



b. $M_\infty = 7.7$

Fig. 12 Concluded

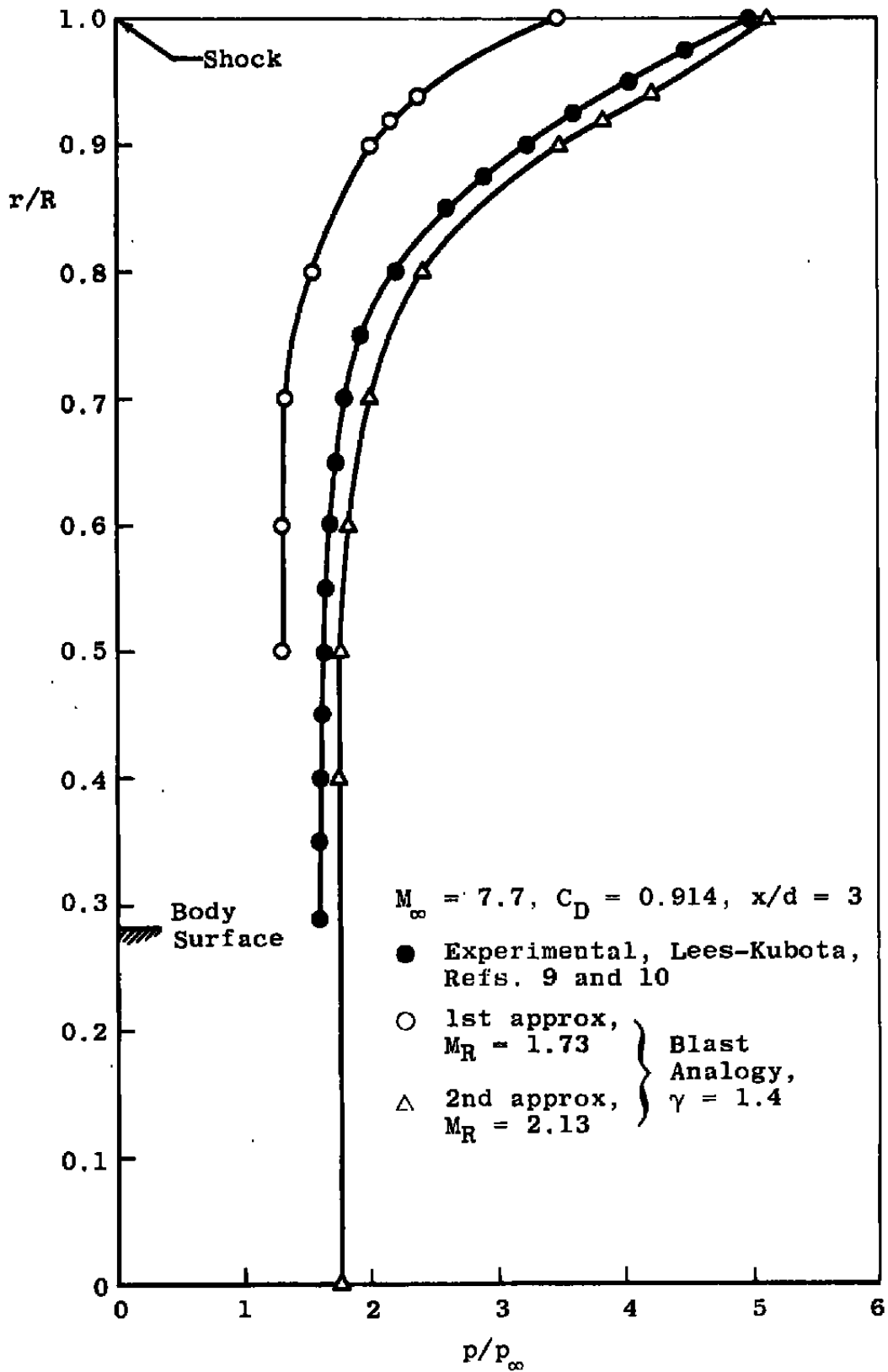


Fig. 13 Radial Pressure Distribution around a Hemisphere Cylinder at $M_\infty = 7.7, x/d = 3$

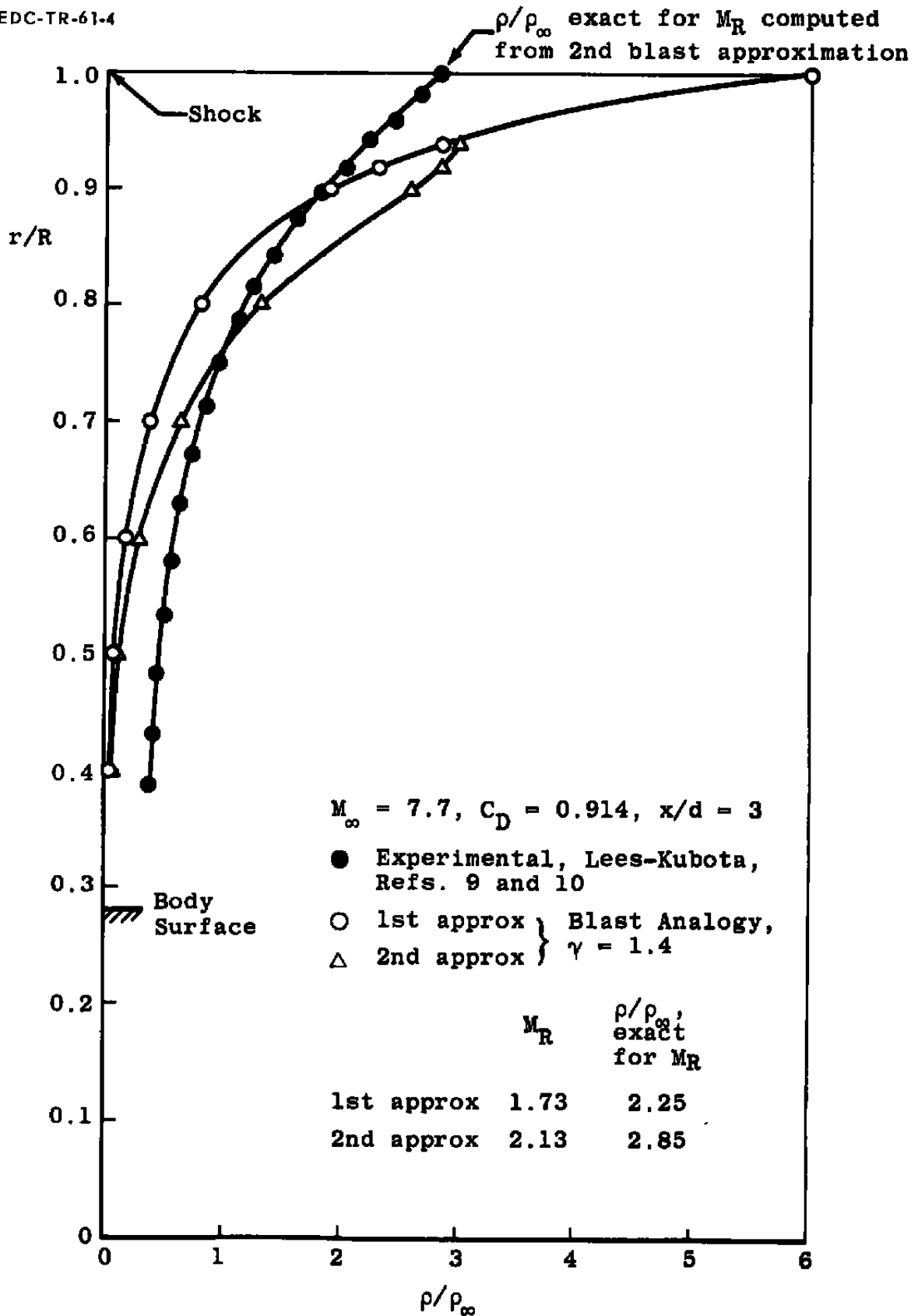


Fig. 14 Radial Density Distribution around a Hemisphere Cylinder at $M_\infty = 7.7, x/d = 3$

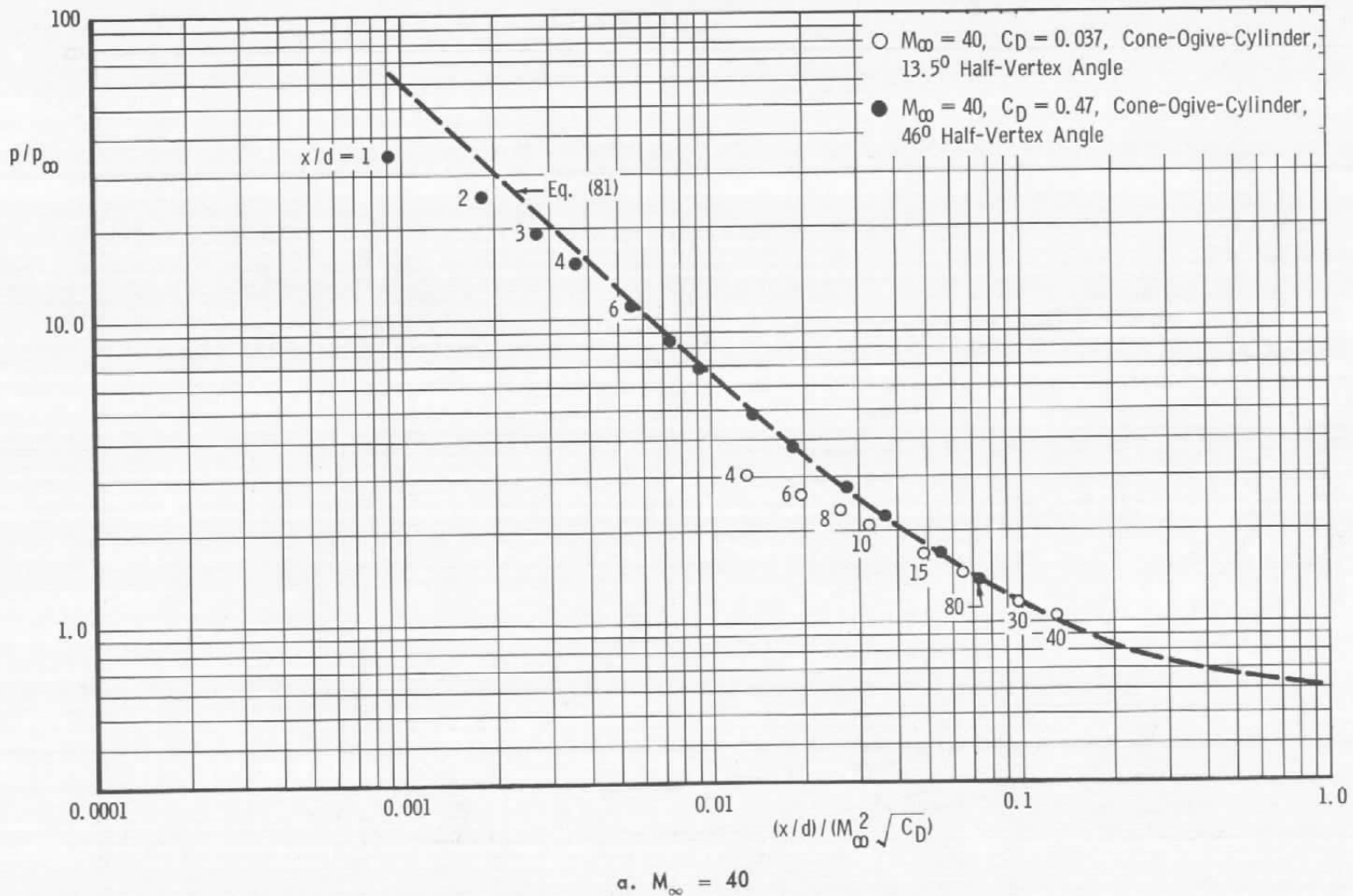
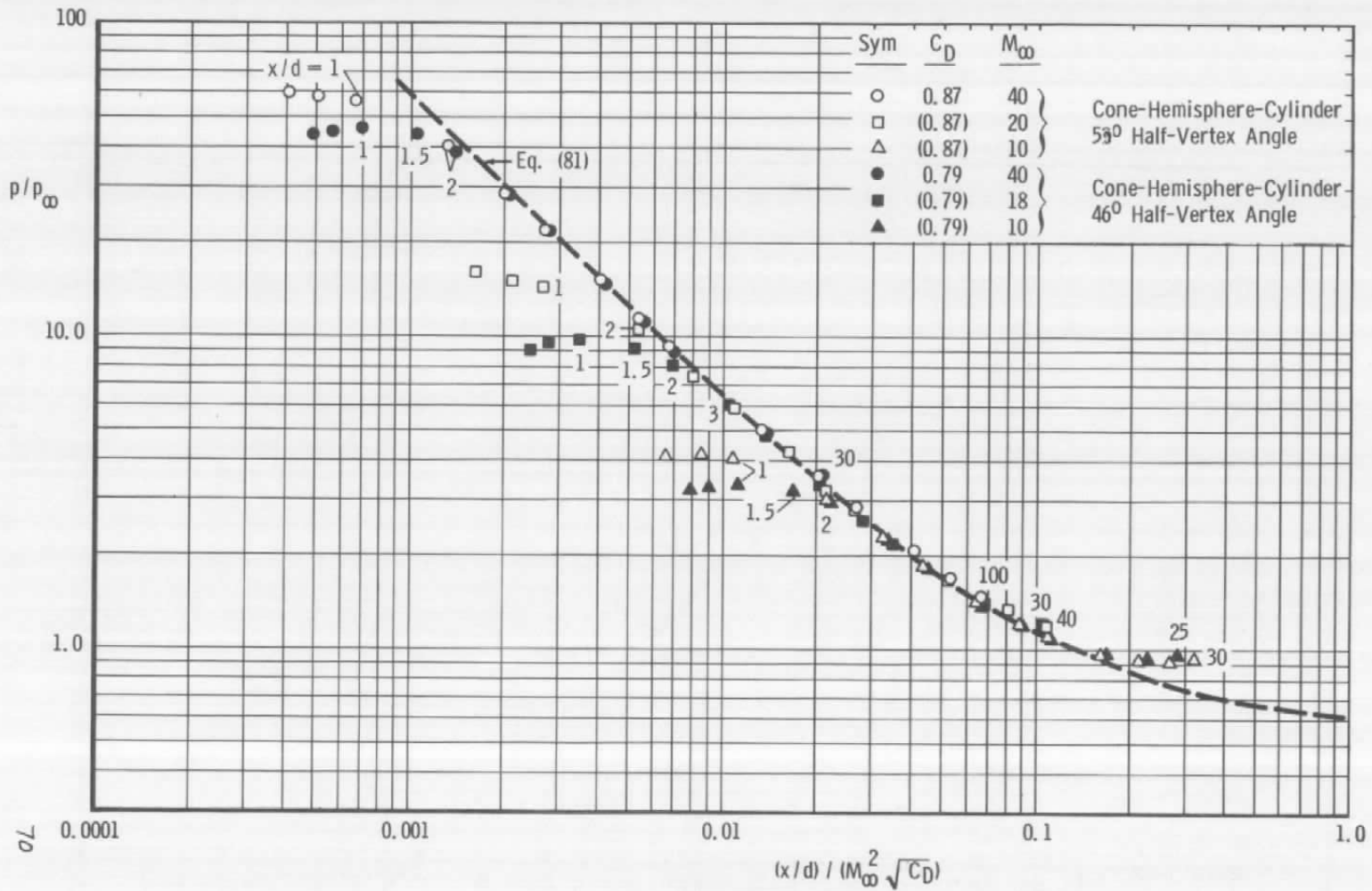
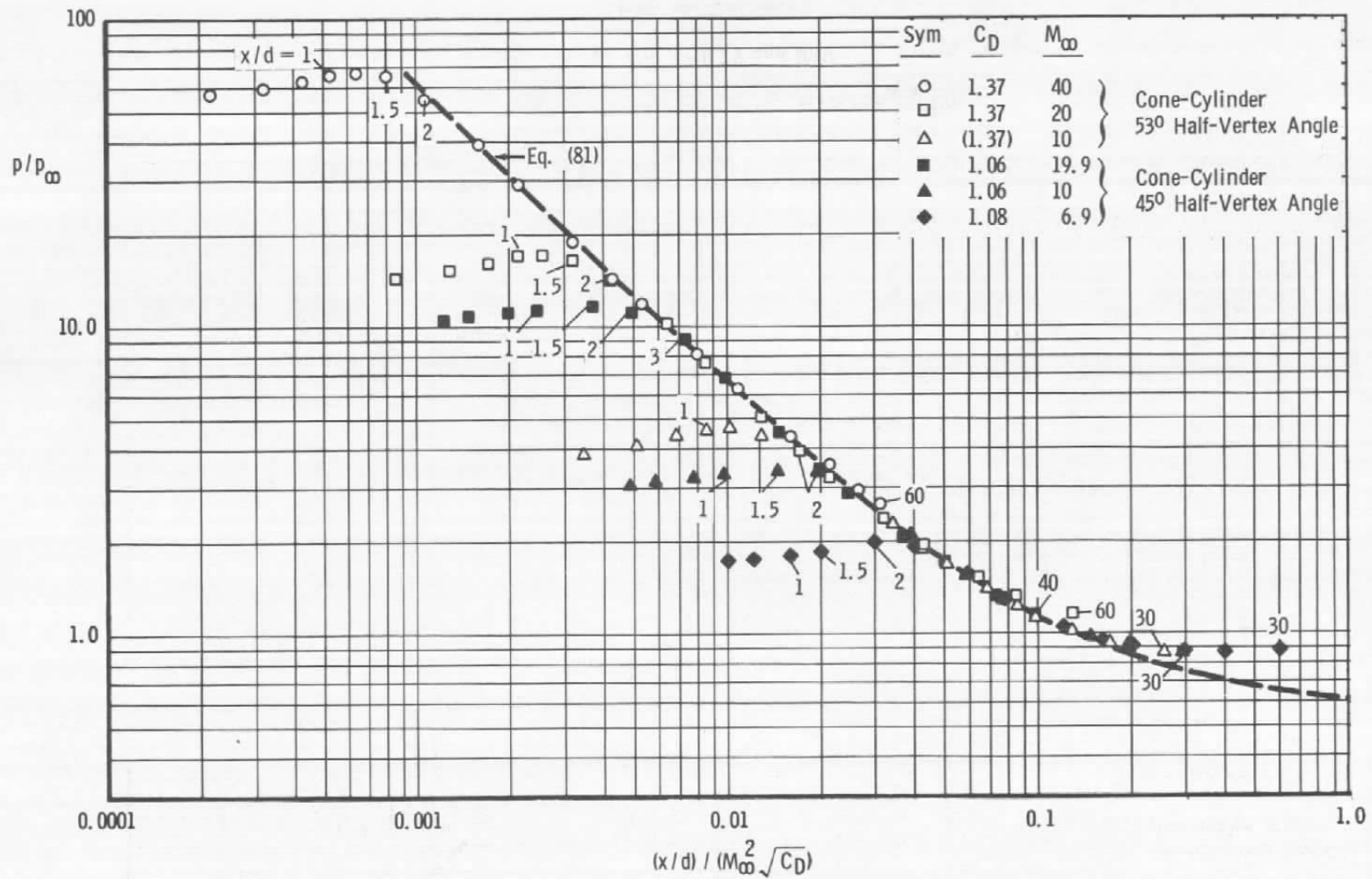


Fig. 15 Correlation of Theoretical Pressure Distribution in Axisymmetric Flow for $\gamma = 1.4$, Ref. 14



b. $C_D = 0.79$ and 0.87

Fig. 15 Continued



c. $C_D = 1.06$ to 1.37

Fig. 15 Concluded

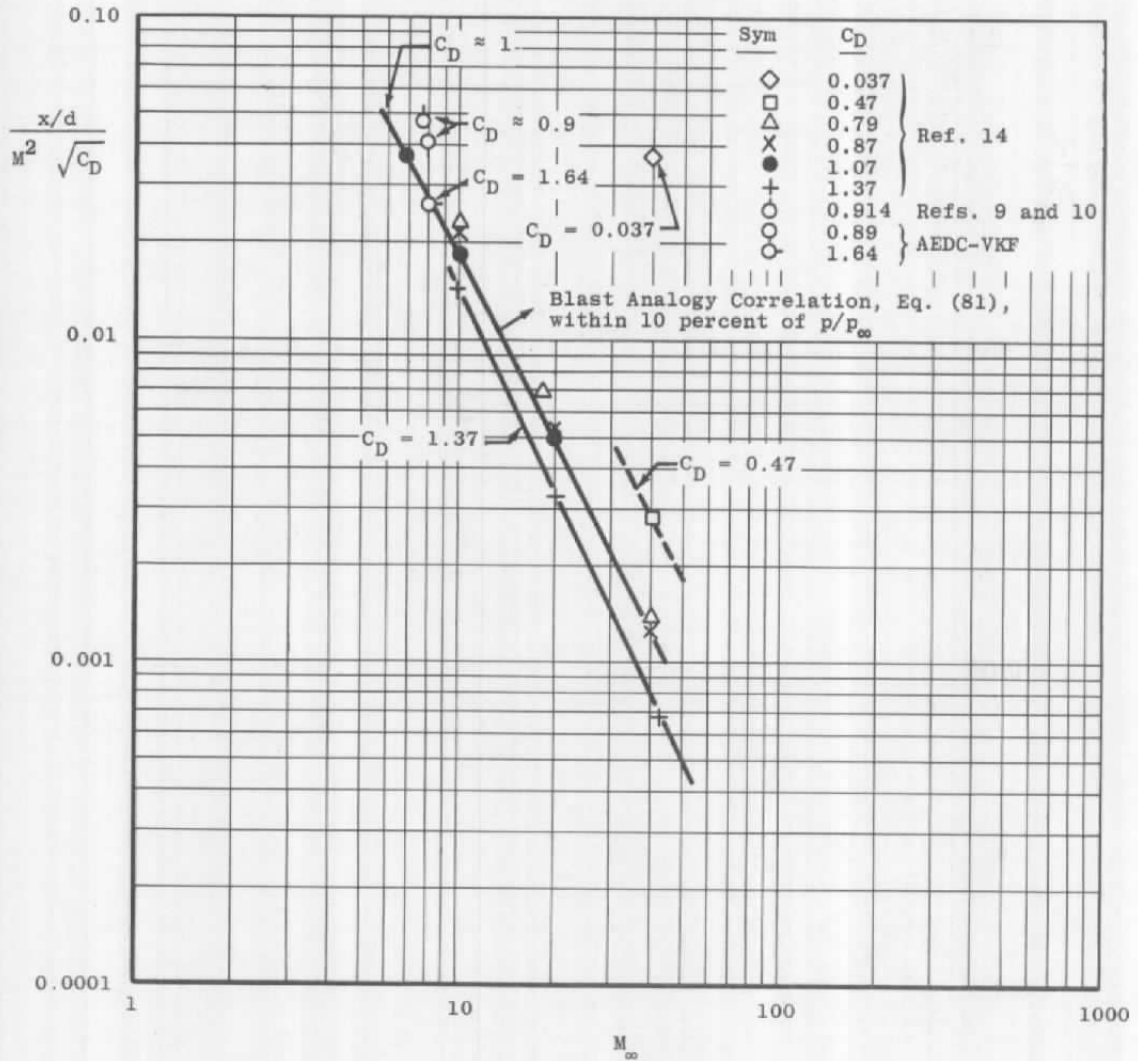


Fig. 16 Limits of Validity of Blast Analogy Correlation for Various Drag Coefficients, $\gamma = 1.4$ (Based on Computations by Van Hise, Ref. 14)

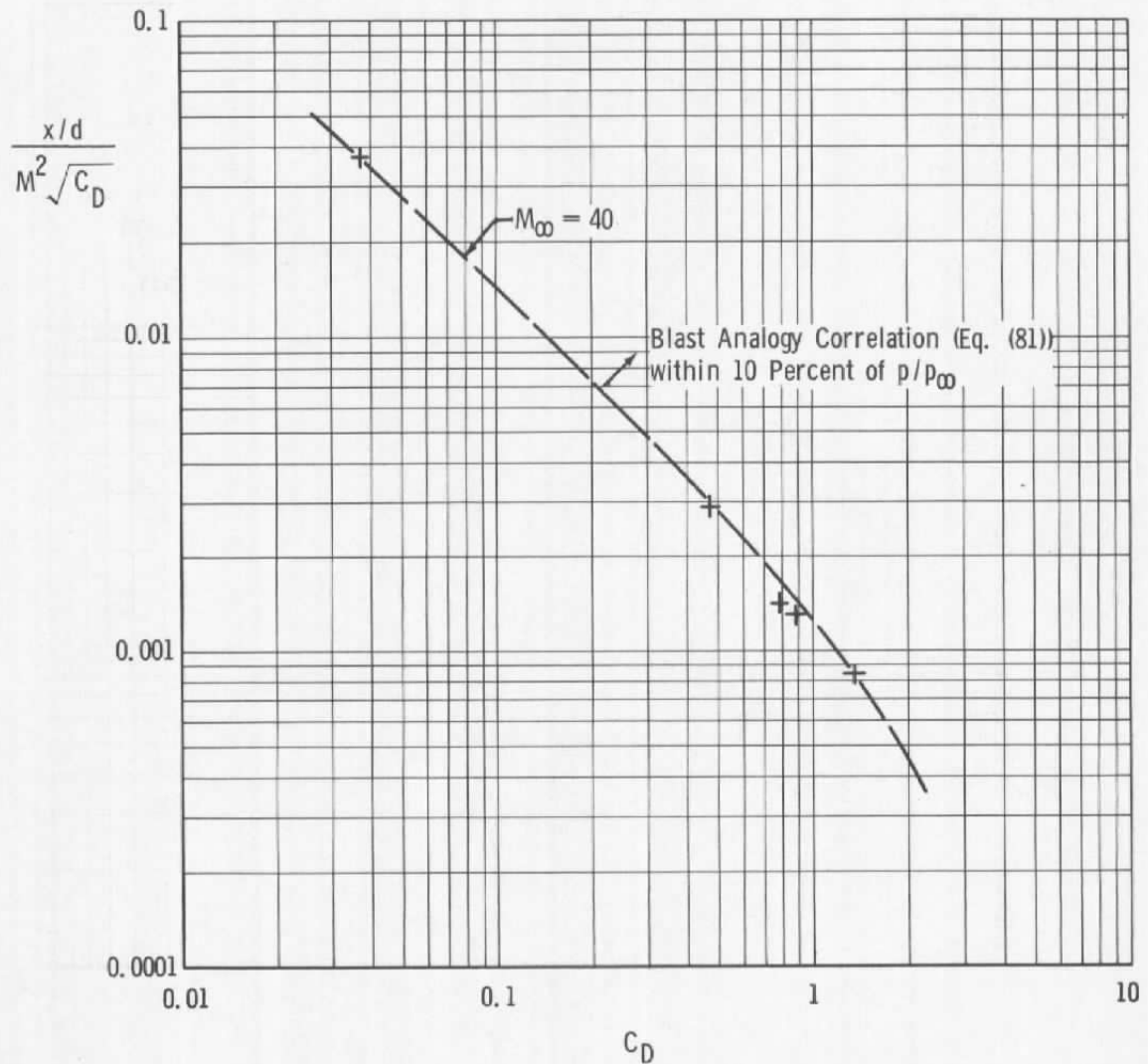


Fig. 17 Limit of Validity of Blast Analogy Correlation at $M_\infty = 40$, $\gamma = 1.4$
(Based on Computations by Van Hise, Ref. 14)

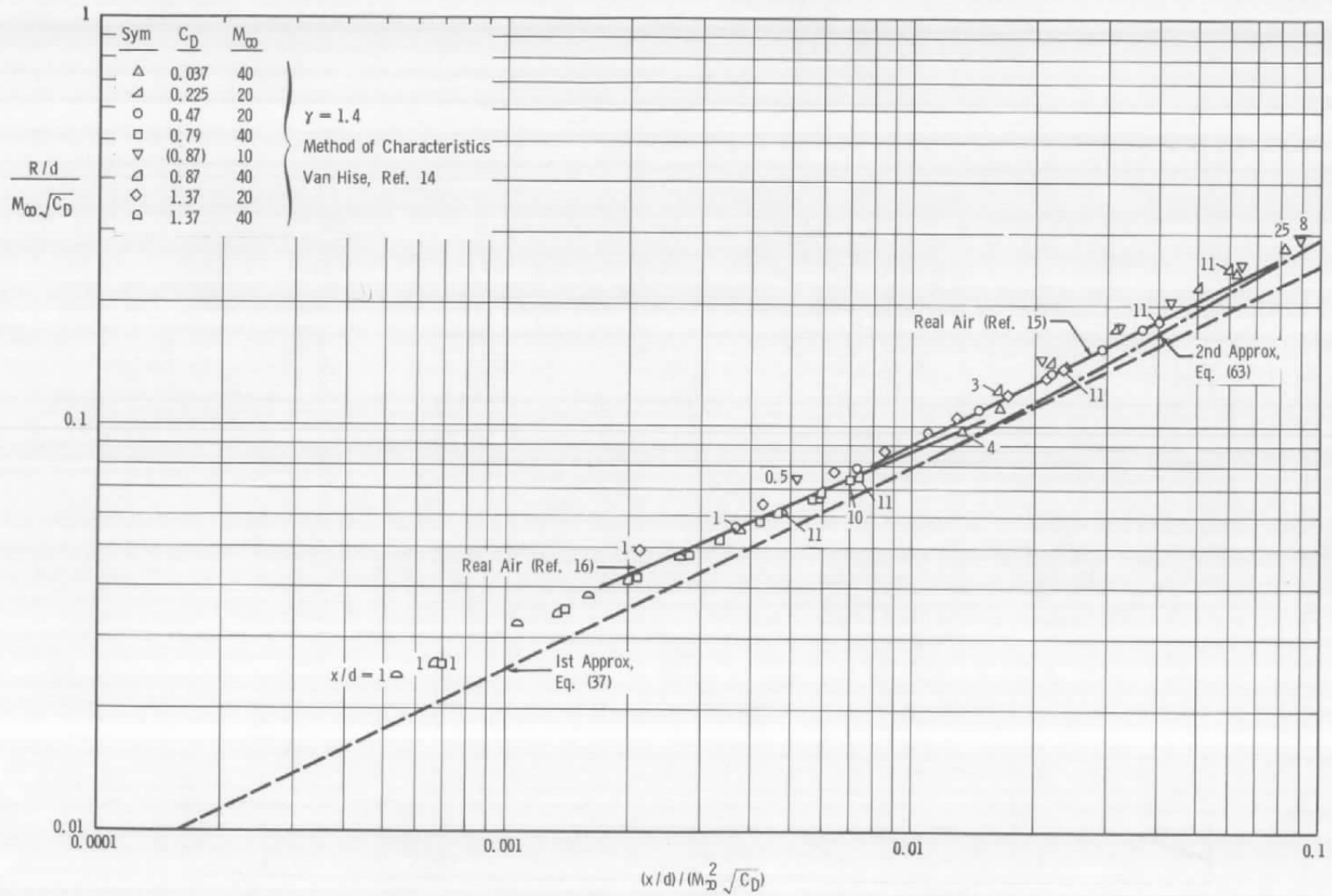
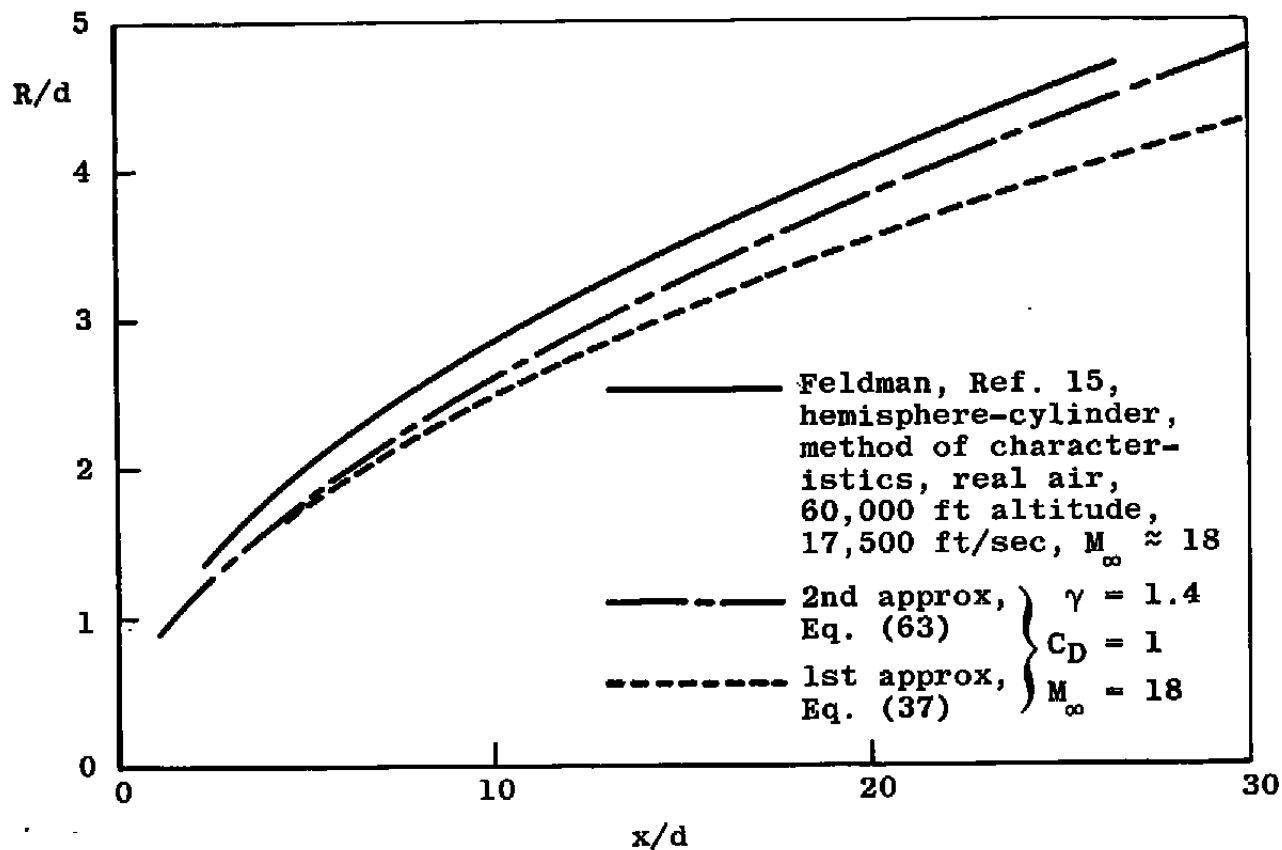


Fig. 18 Correlation of Theoretical (Method of Characteristics, $\gamma = 1.4$) Shock Shapes in Axisymmetric Flow



a. Shock Shapes

Fig. 19 Shock Shapes and Pressure Distributions Calculated by the Method of Characteristics and Blast Analogy in Axisymmetric Flow, $M_\infty = 18$

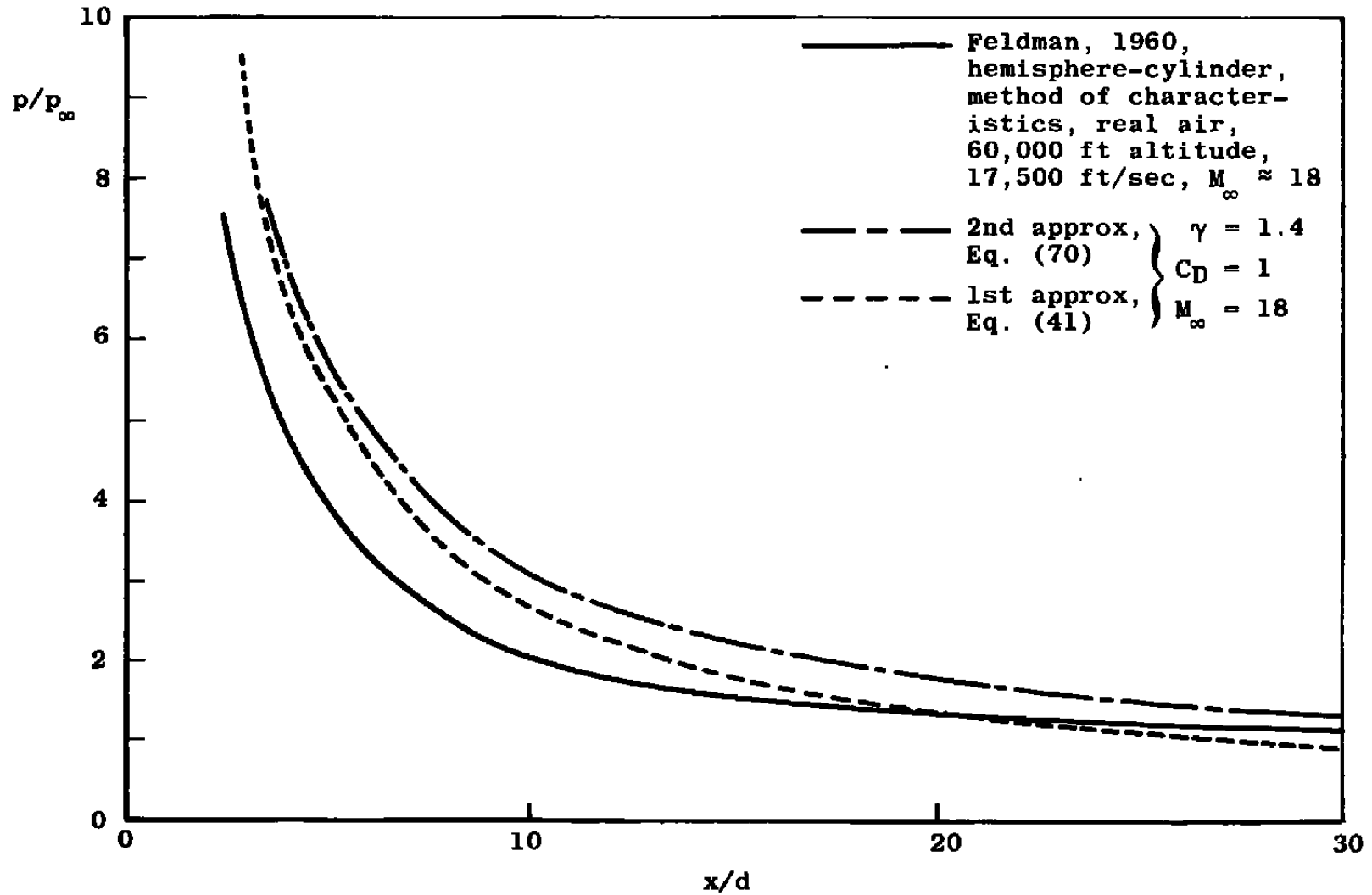
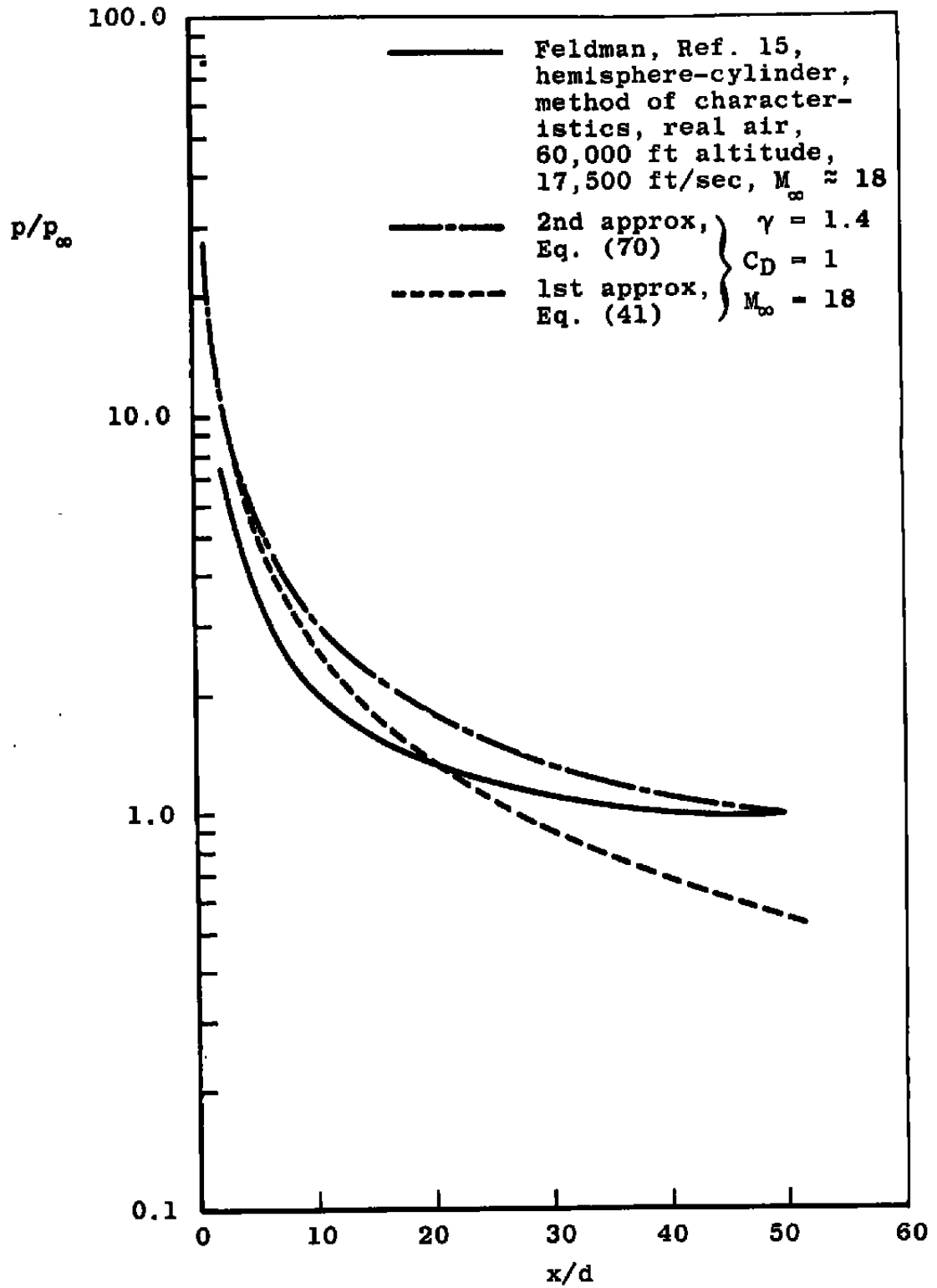
b. Pressure Distributions, $x/d \leq 30$

Fig. 19 Continued



c. Pressure Distributions, $x/d \leq 50$

Fig. 19 Concluded

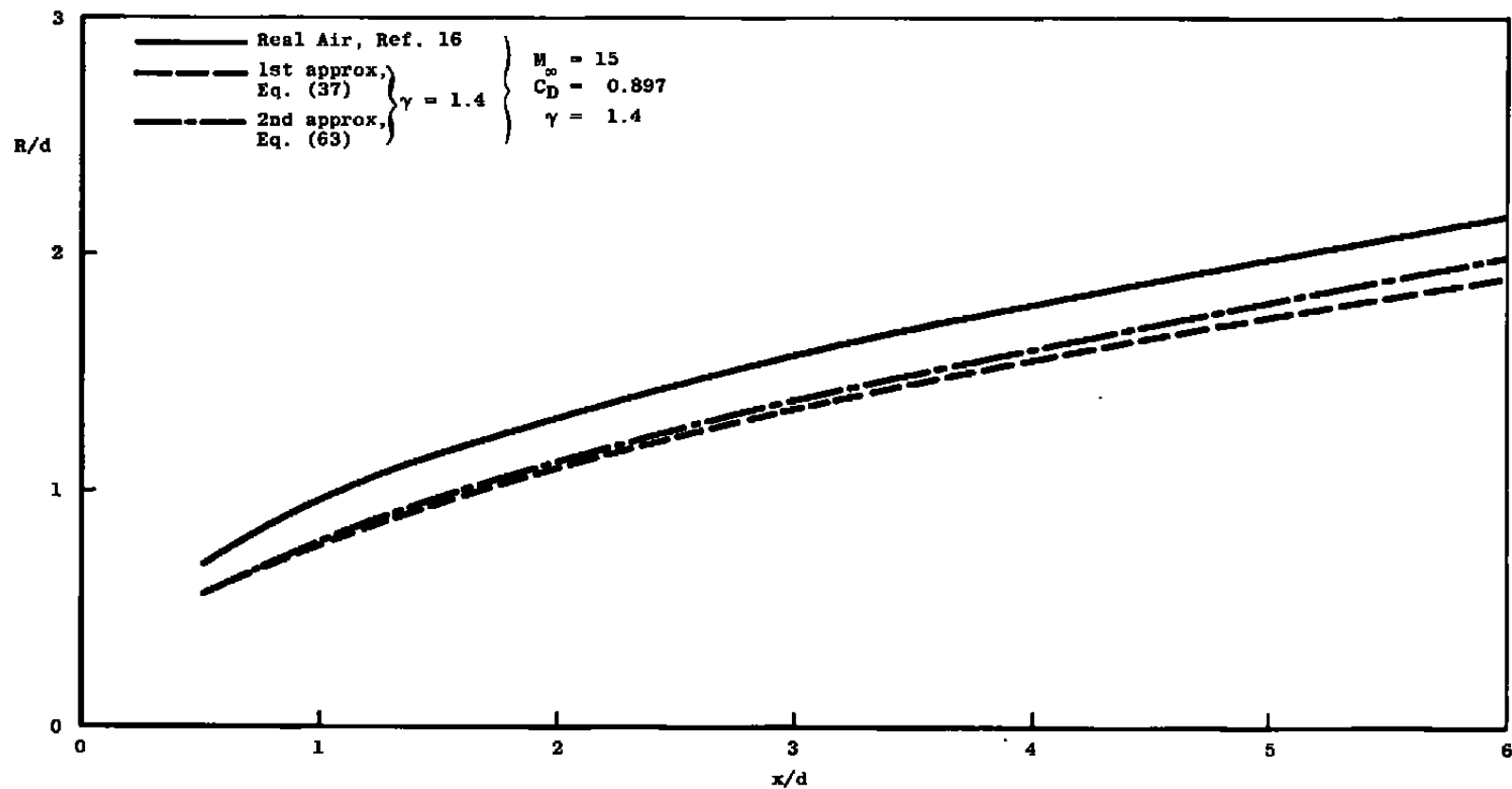


Fig. 20 Shock Shape for a Hemisphere Cylinder at $M_\infty = 15$ in Real Air

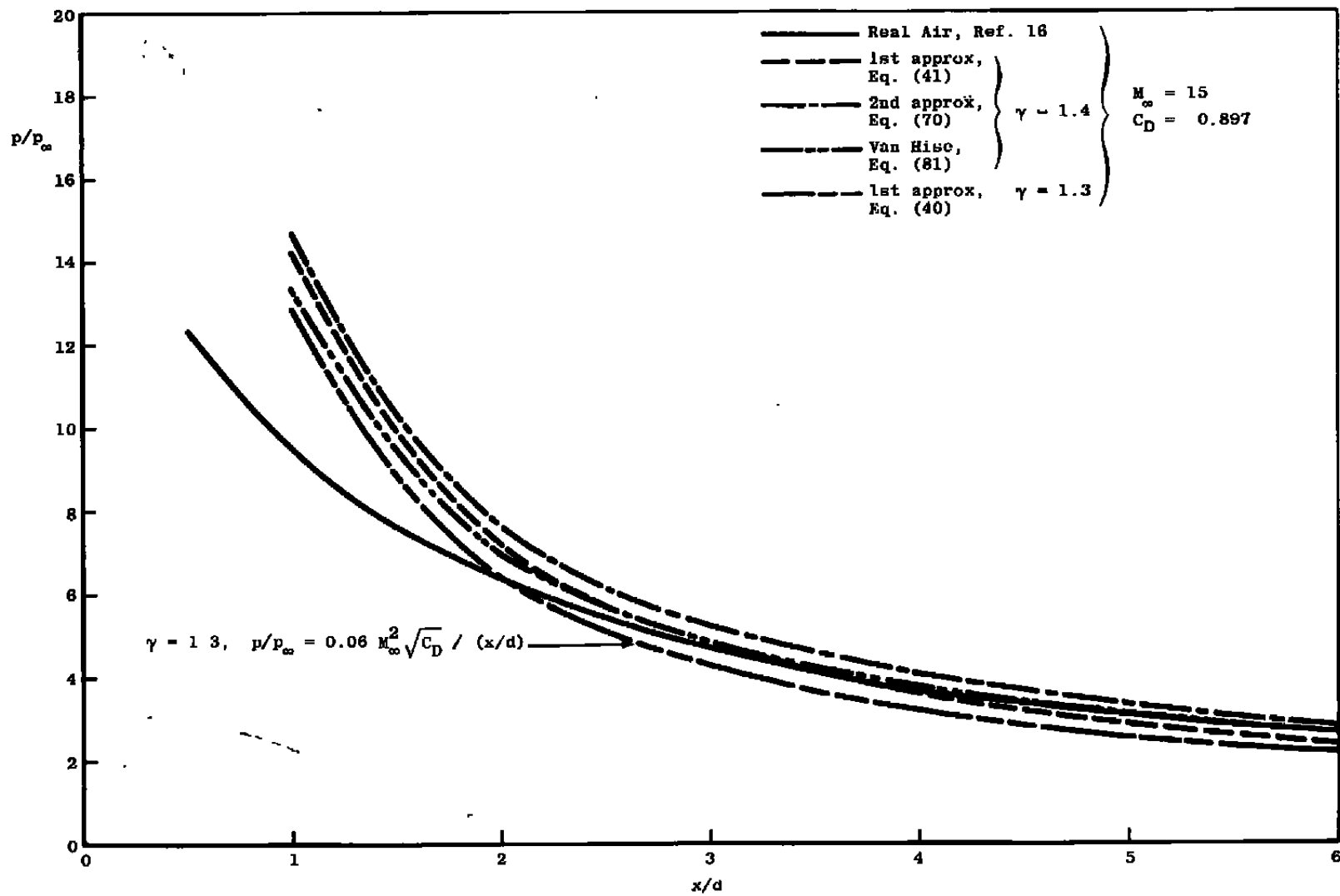


Fig. 21 Pressure Distribution on a Hemisphere Cylinder at $M_{\infty} = 15$ in Real Air

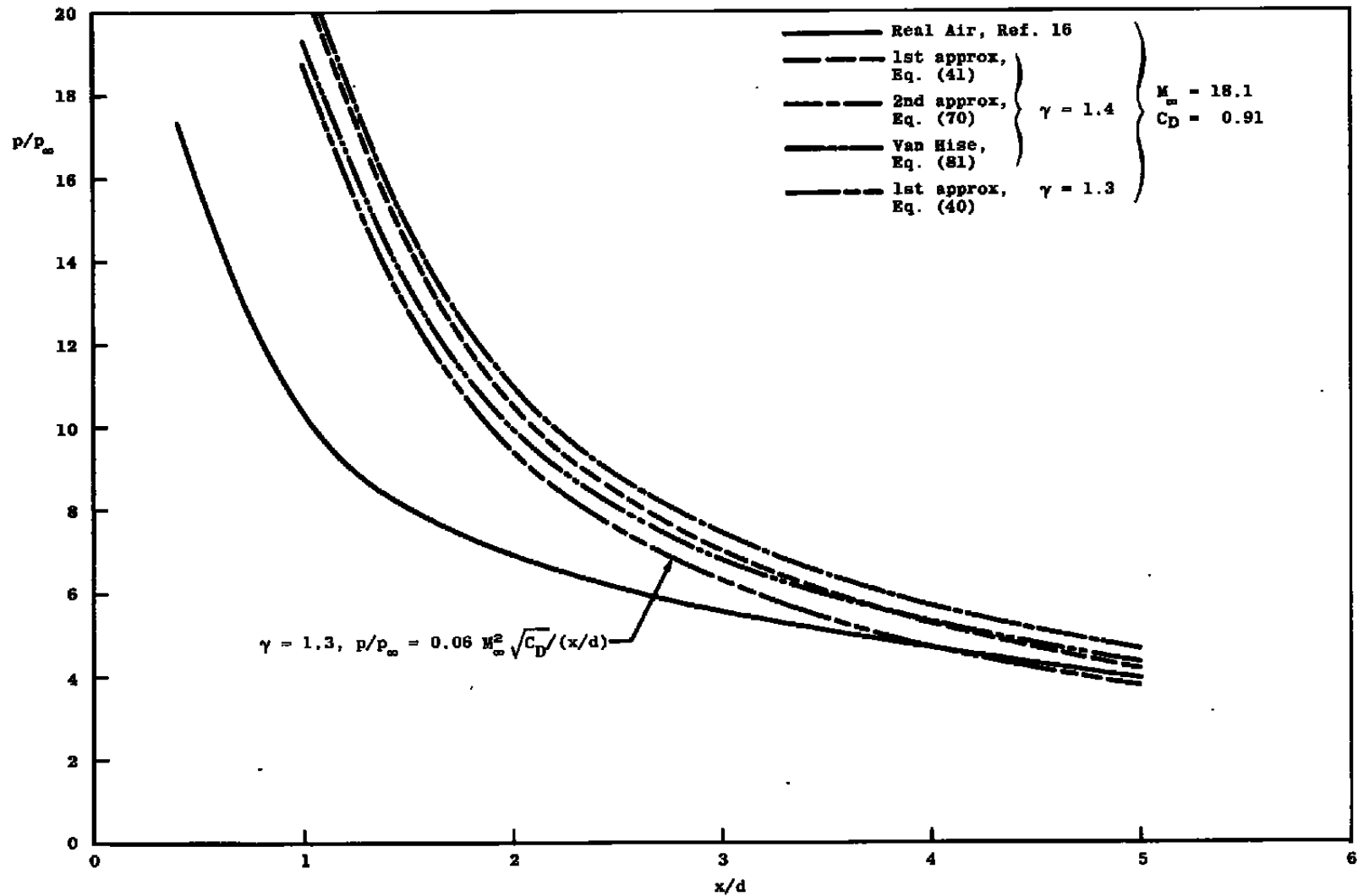


Fig. 23 Pressure Distribution on a Hemisphere Cylinder at $M_\infty = 18.1$ in Real Air

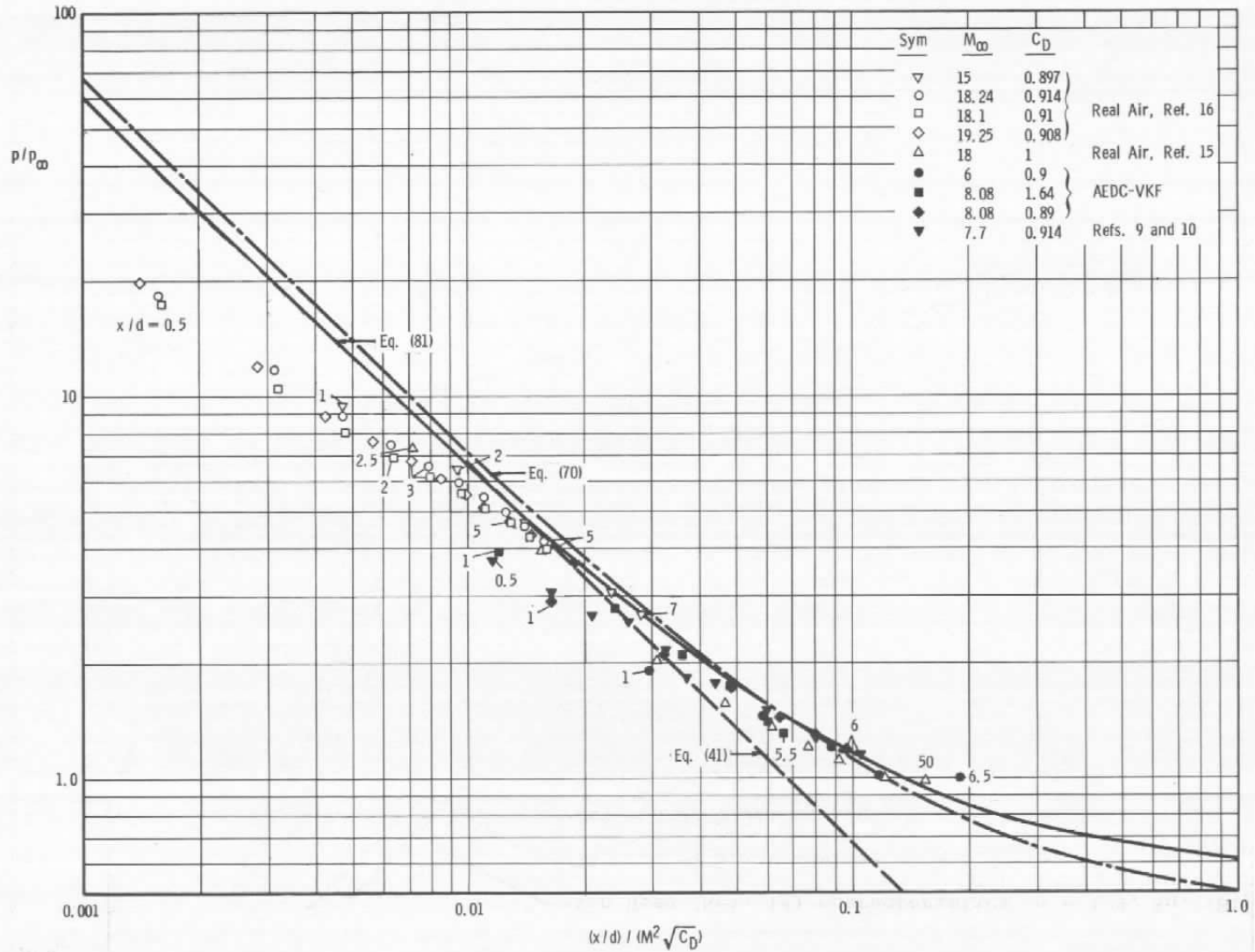


Fig. 24 Correlation of Experimental and Theoretical (for Real Air) Pressure Distribution in Axisymmetric Flow

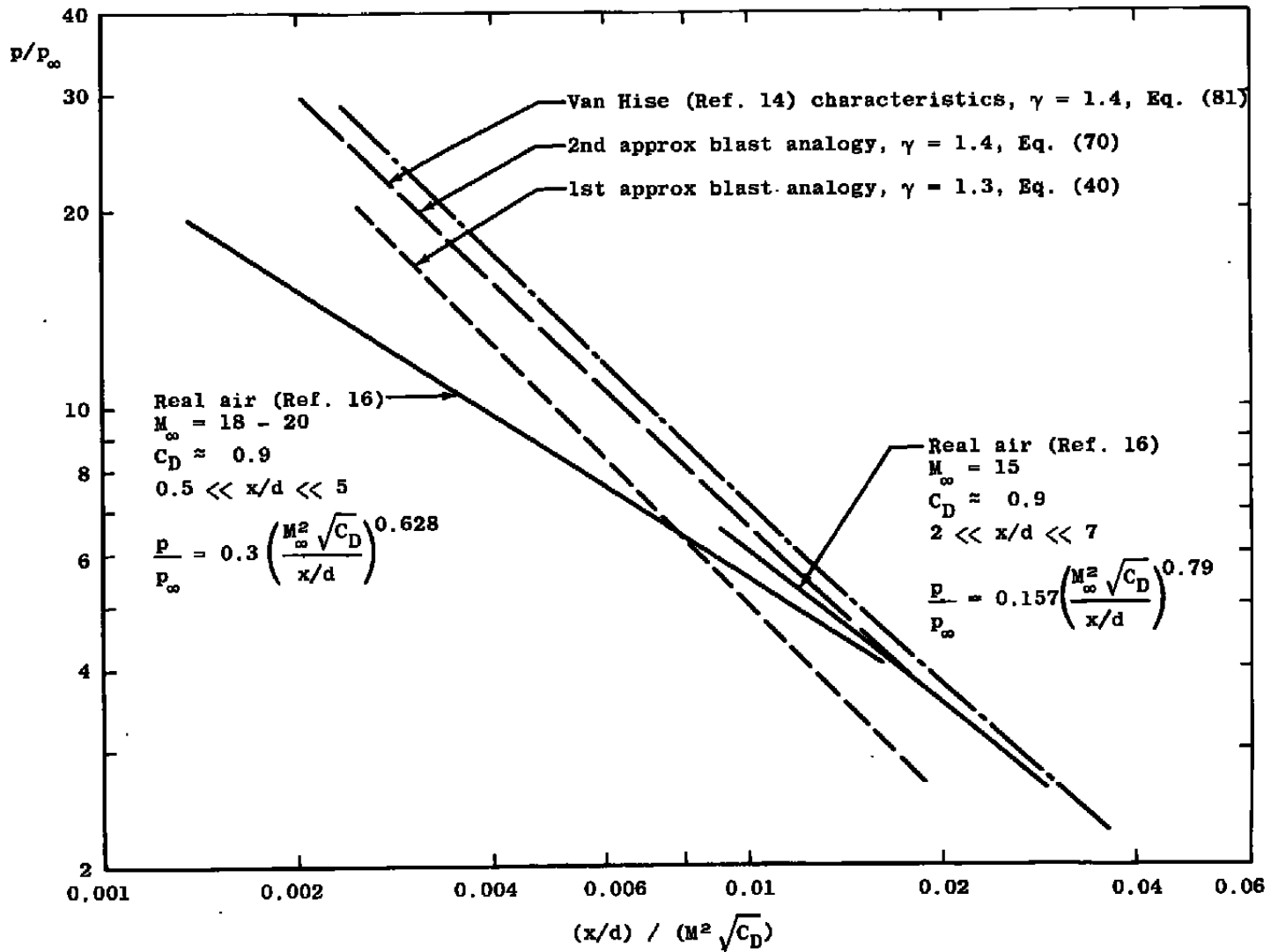


Fig. 25 Correlation of Pressure Distribution in Axisymmetric Flow of Real and Ideal ($\gamma = 1.4$) Air

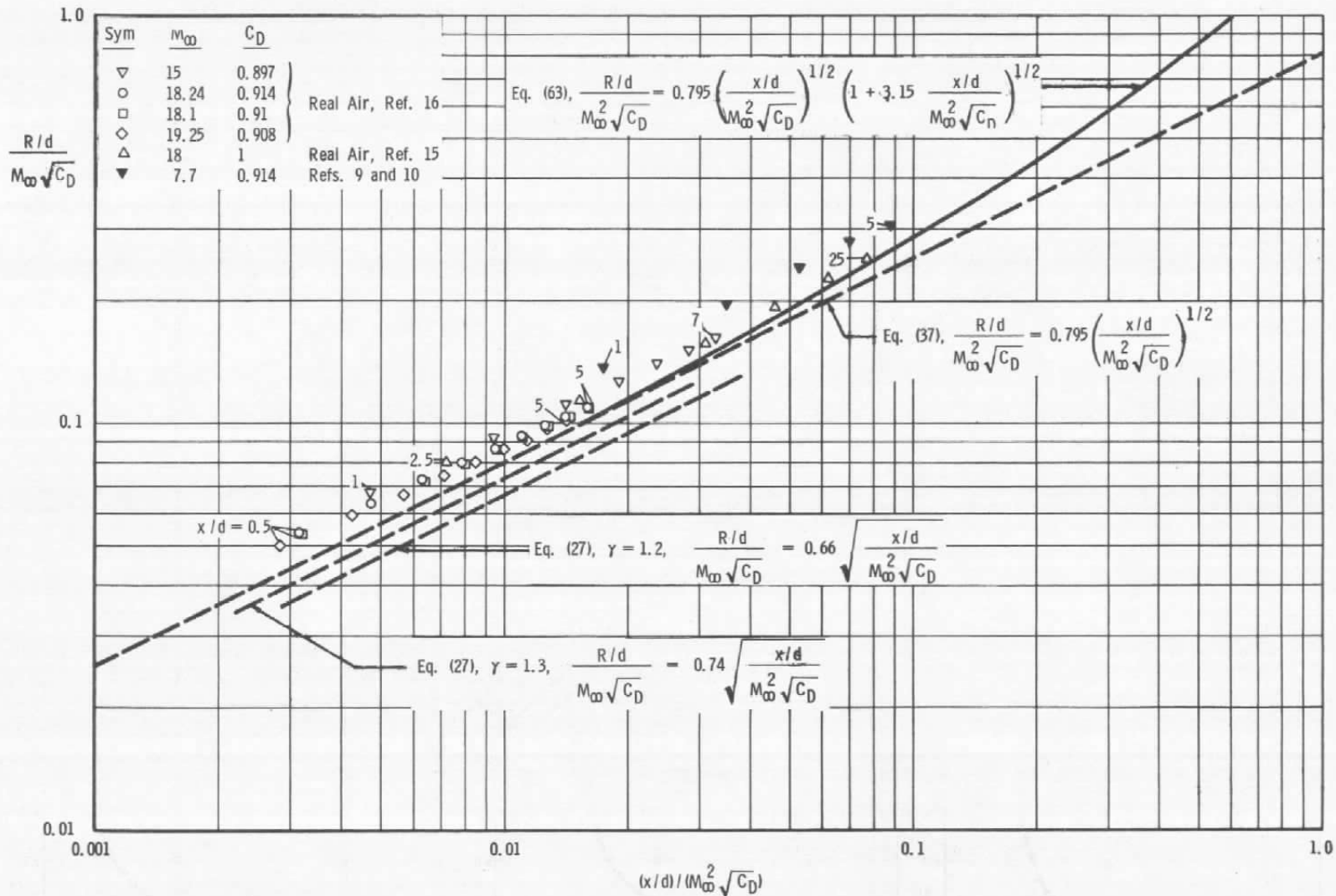


Fig. 26 Correlation of Experimental and Theoretical (for Real Air) Shock Shapes around a Hemisphere Cylinder

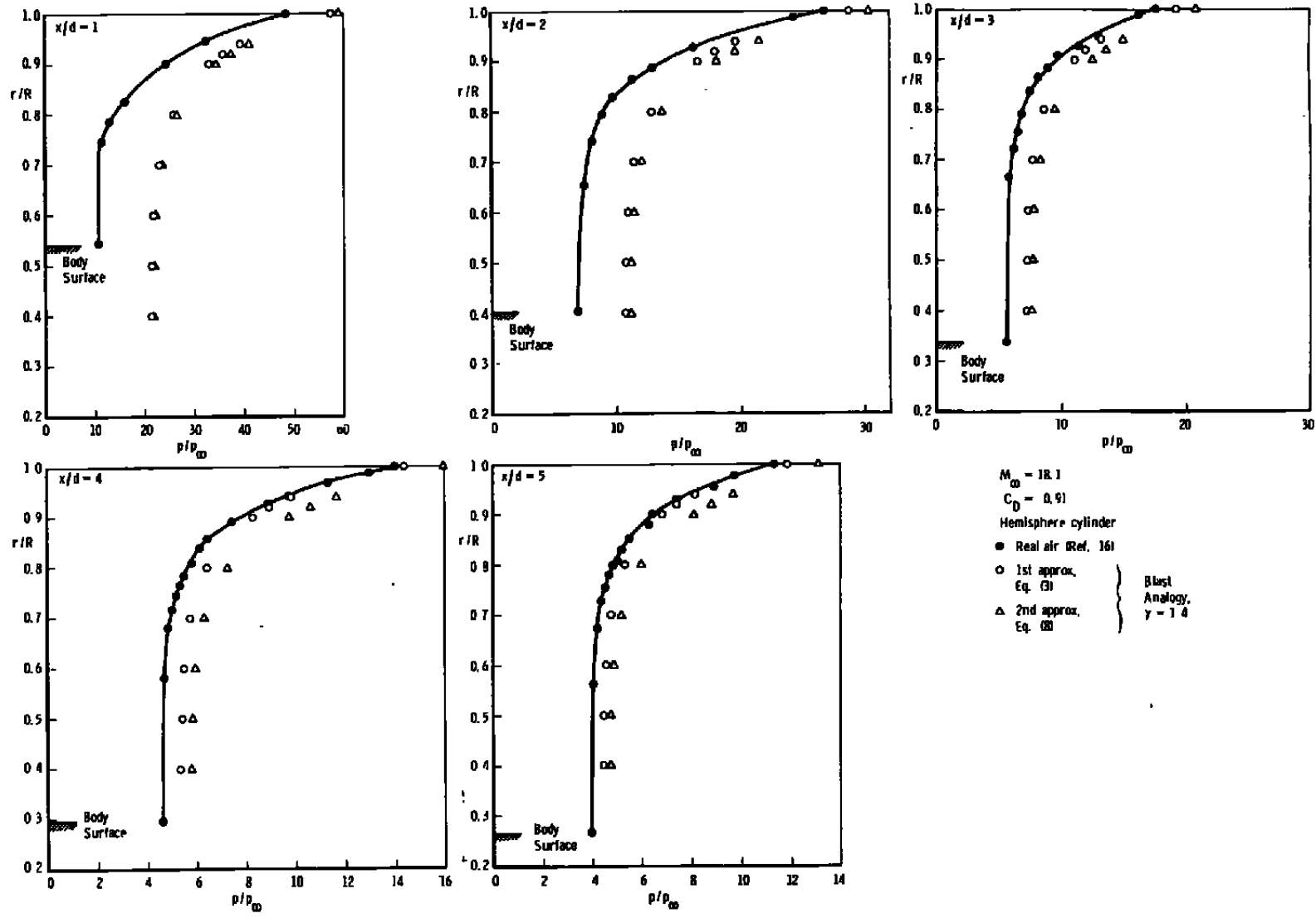


Fig. 27 Radial Pressure Distribution in Flow of Real Air

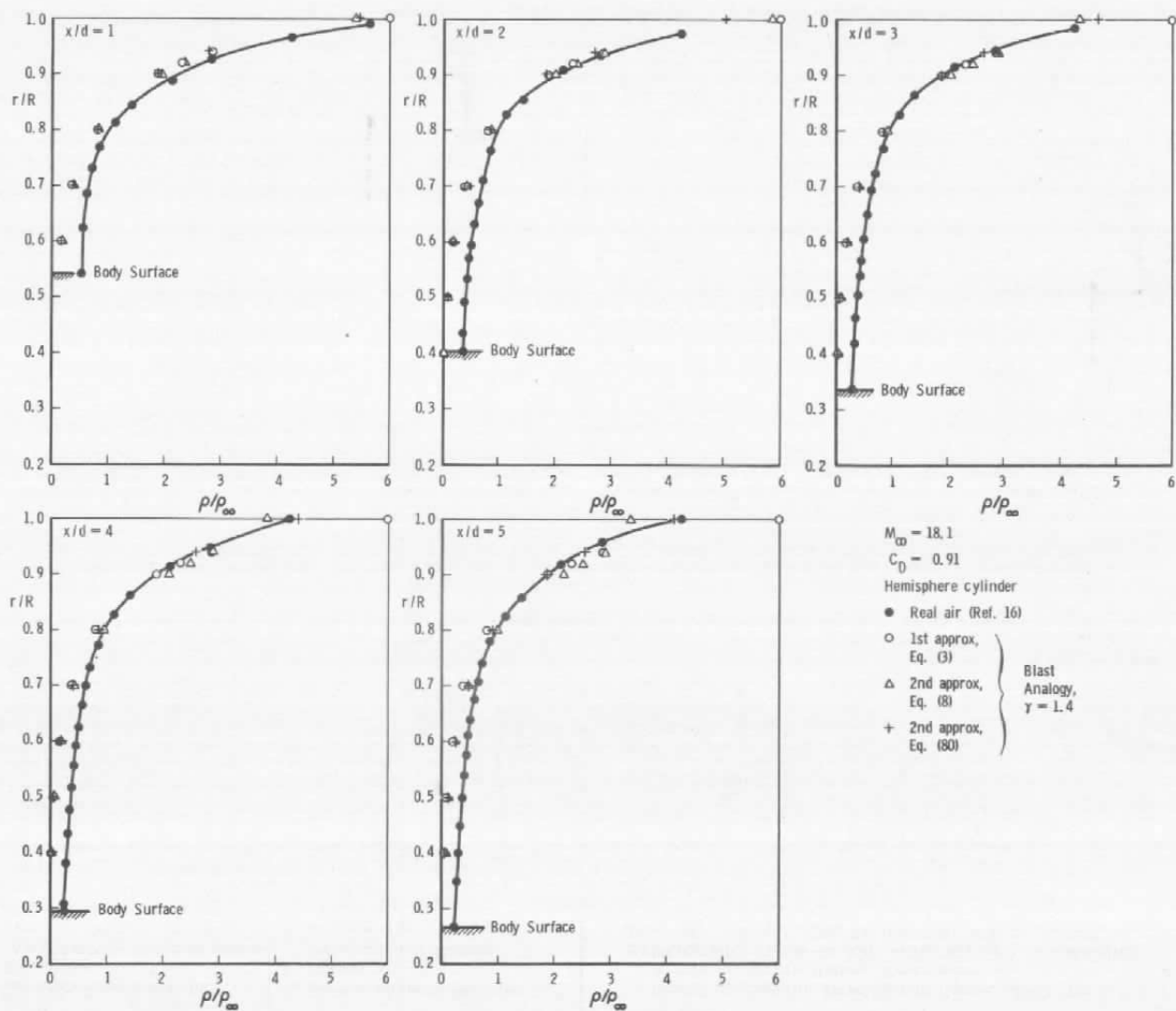


Fig. 28 Radial Density Distribution in Flow of Real Air