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<b>1. REPORT DATE (DD-MM-YYYY)</b> 04/21/2014		<b>2. REPORT TYPE</b> Interim Research Performance Report		<b>3. DATES COVERED (From - To)</b> 1/1/2014 - 3/31/2014	
<b>4. TITLE AND SUBTITLE</b> Interim Research Performance Report Quarterly Report No. 10				<b>5a. CONTRACT NUMBER</b>	
				<b>5b. GRANT NUMBER</b> N00014-11-1-0752	
				<b>5c. PROGRAM ELEMENT NUMBER</b>	
<b>6. AUTHOR(S)</b> Nathan E. Murray, Charles E. Tinney (U. of Texas at Austin), Praveen Panickar (CRAFT Tech.)				<b>5d. PROJECT NUMBER</b>	
				<b>5e. TASK NUMBER</b>	
				<b>5f. WORK UNIT NUMBER</b>	
<b>7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES)</b> The University of Mississippi Jamie Whitten National Center for Physical Acoustics University, MS 38677				<b>8. PERFORMING ORGANIZATION REPORT NUMBER</b>	
<b>9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES)</b> Joseph Doychak Office of Naval Research 875 North Randolph Street Arlington, VA 22203-1995				<b>10. SPONSOR/MONITOR'S ACRONYM(S)</b> ONR	
				<b>11. SPONSOR/MONITOR'S REPORT NUMBER(S)</b>	
<b>12. DISTRIBUTION/AVAILABILITY STATEMENT</b> Approved for Public Release; Distribution is Unlimited					
<b>13. SUPPLEMENTARY NOTES</b>					
<b>14. ABSTRACT</b> Results from the analysis of the acoustic waveforms produced by a cold Mach 3.0 jet were compiled and submitted for publication. Efforts now turn to completing laboratory experiments on the heated, conic-section nozzle, supersonic jet in the U. Miss. facility. Initial shadowgraph images provide insight into the generation of shock-like events in the jet's near-field.					
<b>15. SUBJECT TERMS</b> Jet Noise Reduction, High Dynamic Range PIV, Computational Phased Array Beamforming, Aeroacoustics					
<b>16. SECURITY CLASSIFICATION OF:</b>			<b>17. LIMITATION OF ABSTRACT</b>	<b>18. NUMBER OF PAGES</b>  41	<b>19a. NAME OF RESPONSIBLE PERSON</b> Nathan E. Murray
<b>a. REPORT</b>	<b>b. ABSTRACT</b>	<b>c. THIS PAGE</b>			<b>19b. TELEPHONE NUMBER (Include area code)</b> 662-915-3190

TOWARD ACTIVE CONTROL OF NOISE  
FROM HOT SUPERSONIC JETS

## Quarterly Progress Report No. 10

1 JANUARY 2014 – 31 MARCH 2014

Nathan E. Murray (PI)  
National Center for Physical Acoustics  
The University of Mississippi  
University, MS 38677  
(662) 915-3190  
nmurray@olemiss.edu

Charles E. Tinney (Co-PI) – The University of Texas at Austin  
Praveen Panickar – Combustion Research and Flow Technology, Inc.

Contract: N00014-11-1-0752

Attn: Joseph Doychak & Brenda Henderson  
Office of Naval Research  
Arlington, VA 22203-1995  
joseph\_doychak@onr.mil | brenda.s.henderson@navy.mil

### Executive Summary

Results from the analysis of the acoustic waveforms produced by a cold Mach 3.0 jet were compiled and submitted for publication. Efforts now turn to completing laboratory experiments on the heated, conic-section nozzle, supersonic jet in the U. Miss. facility. Initial shadowgraph images provide insight into the generation of shock-like events in the jet's near-field.

20140428077

April 21, 2014

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## 1 Project Objectives and Status

### 1.1 Review of Program Objectives

Broadly, the objectives of this effort focus on the finite amplitude disturbances generated by supersonic jets, their impact on measurement and modeling of the jet noise, and the importance of these events in evaluation of any jet noise control strategy. Experimental measurements have been obtained for shock-free and shock-containing supersonic jets.

- An acoustic survey of the sound field indicates that the sound source is in the post-potential core region. Contours of constant amplitude are curved indicating that a distribution of convection velocities is associated with the source disturbances.
- Evaluation of the “shock” content in the pressure time-series indicates finite amplitude events can be observed immediately in the near-field as well as being the result of steepening in the mid-field.
- Evaluation of the decay along the peak emission angle indicates a transition from cylindrical to spherical spreading occurs between 12 and 33 jet diameters in radius.
- Evaluation of the time series of the pressure along the peak emission angle indicates some waves may coalesce while others may steepen.
- Shadowgraph images verify finite amplitude disturbances are emitted both from the near-nozzle flow and the post-potential core region.

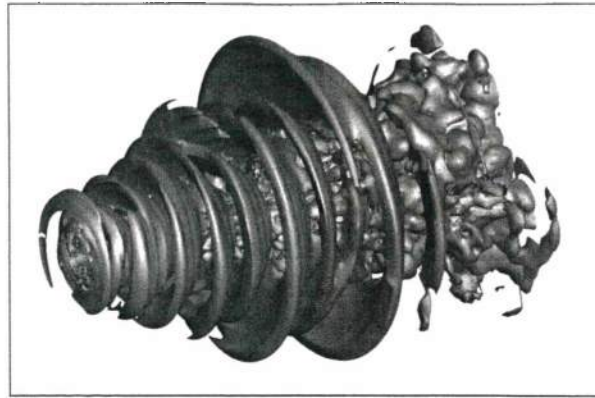


Figure 1. The acoustic disturbances in the near-field of the jet appear as conical-shaped, expanding surfaces. Flow is left-to-right.

## 1.2 Project Status

The results of the most recent set of measurements conducted in the UT Austin facility have been submitted to the *AIAA Journal* for publication. Now, the focus of the remaining experimental efforts shift to the U. of Miss. facility where acoustic and flow measurements will be conducted with a hot-jet whose acoustic Mach number is higher than that of the cold jet used for the UT Austin measurements.

## 2 Activity for Current Reporting Period

### 2.1 Disturbances in the Near-Field of the Jet

Iso-surfaces of the density were analyzed from the large-eddy simulation (LES) of the over-expanded, conic-section nozzle ( $NPR = 3.96$ ,  $TTR = 3.35$ ,  $M_j = 1.55$ ,  $M_a = 2.34$ ). The near-field structure is characterized by conical shaped surfaces as illustrated in Figure 1. The cone half-angle of these structures is associated with the convection velocity of the underlying disturbance. Convection speeds between 0.7-0.8 of the jet's acoustic Mach number suggest angles between 32 and 37 degrees. However, this velocity is not constant along the jet axis.

A contour plot of the dilatation from the LES is shown in Figure 2. The annotations (black lines) highlight the angle that the disturbance appears to make. These angles are steeper at the jet exit and flatten out in the downstream direction. This indicates that the underlying disturbances must be slowing down as they move along the jet. Certainly, the far-field directivity has a particular peak, but in the near-field the underlying disturbances should be considered to result from some distribution of convection speeds.

Shadowgraph images were recorded in the near-field of both the nozzle exit and post-potential core regions of the jet. A retro-reflective shadowgraph setup was used to record the images. The near-nozzle region exhibits a large number of shock-like disturbances being generated. While not regular, the images recorded in the post-potential core region also exhibit a stochastic occurrence of finite amplitude disturbances. The shape and direction of these disturbances suggest that they originated in the post-potential core region of the jet.

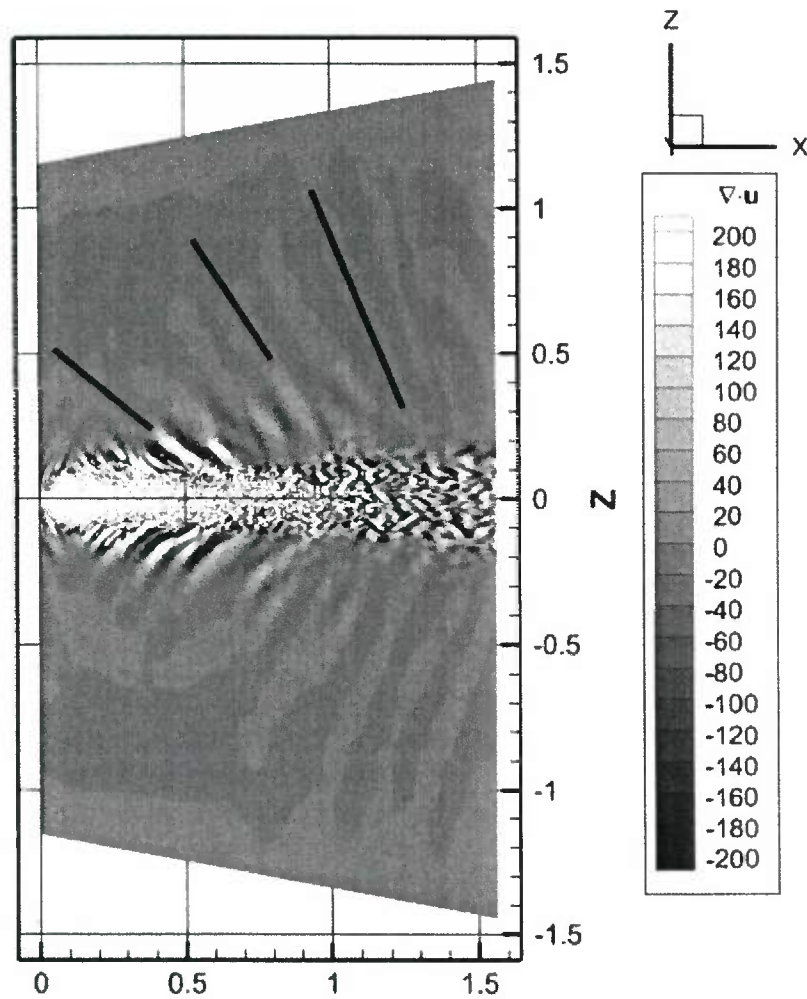
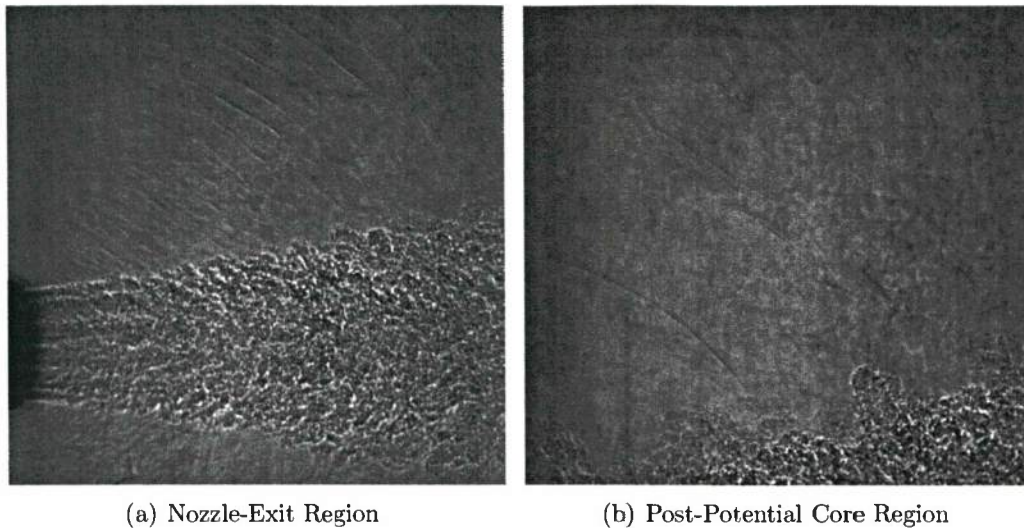


Figure 2. Instantaneous dilatation ( $\nabla \cdot \vec{u}$ ) field computed from the LES of the over-expanded jet.





**Figure 3. Shadowgraph imagery showing finite amplitude disturbances emanating from both the near-nozzle region and the post-potential core region. The vertical line in (b) is located 8 jet diameters downstream from the nozzle exit.**

This observation suggests that there is a distribution of disturbances along the jet that reach supersonic phase speeds with sufficient amplitude to generate finite amplitude waves in the near-field of the jet.

## 2.2 Wave Coalescence

An inspection of Figure 3(a) indicates that the shock-like structures are coalescing as they propagate away from the jet. This supports concepts suggested by the acoustic measurements performed by the UT Austin group (see Appendix A). This process is only well understood in a 1-D sense. The 3-dimensional merging of finite amplitude disturbances is not well understood, and the effect, if any, on the jet's near- and far-field directivity is unknown at this time.

## 3 Technical/Cost Status & Problem Areas

None.

## 4 Publications, Meetings, and/or Travel

- Dr. Tinney's student, Romain Fiévet, presented recent experimental results at the 2014 Texas Fluid Dynamics Conference. The abstract is included in Appendix B.
- Results from recent work performed by Dr. Tinney and his students have been submitted to the *AIAA Journal*. The submission draft is included in Appendix A.

- Dr. Tinney's former student, Dr. Baars, completed a journal submission to *Physics of Fluids* on the "POD-Based Spectral Higher-Order Stochastic Estimation." This concept was included in the original proposal for this project, so it is mentioned here and included in the publications list.

#### 4.1 Running List of Publications Produced

- [1] BAARS, W.J., TINNEY, C.E., MURRAY, N.E., JANSEN, B.J. & PANICKAR, P. (2011) "The Effect of Heat on Turbulent Mixing Noise in Supersonic Jets," in *49<sup>th</sup> AIAA Aerospace Sciences Meeting*, Paper 2011-1029 (Orlando, FL).
- [2] BAARS, W.J., TINNEY, C.E. & WOCHNER, M.S. (2012) "Nonlinear Noise Propagation from a Fully Expanded Mach 3 Jet," in *50<sup>th</sup> AIAA Aerospace Science Meeting*, Paper 2012-1177 (Nashville, TN).
- [3] MURRAY, N.E., LYONS, G.W., TINNEY, C.E., DONALD, B., BAARS, W.J., THUROW, B.S., HAYNES, R.H. & PANICKAR, P. (2012) "A Laboratory Framework for Synchronous Near/Far-Field Acoustics and MHz PIV in High-Temperature, Shock-Containing, Jets," in *Proceedings of the Internoise 2012/ASME NCAD Meeting*, ASME/NCAD-1270 [invited] (New York City, NY).
- [4] BAARS, W.J. & TINNEY, C.E. (2012) "Scaling Model for Nonlinear Supersonic Jet Noise," in *Bulletin of the American Physical Society*, **57**:17, Abstract D24:8 (San Diego, CA).
- [5] PANICKAR, P., ERWIN, J., SINHA, N., MURRAY, N.E. & LYONS, G.W. (2013) "Localization of Acoustic Sources in Shock-Containing Jet Flows Using Phased Array Measurements," in *51<sup>st</sup> AIAA Aerospace Sciences Meeting*, 2013-0613 (Grapevine, TX).
- [6] HAYNES, R.H., BROCK, B.A. & THUROW, B.S. (2013) "Application of MHz Frame Rate, High Dynamic Range PIV to a High-Temperature, Shock-Containing Jet," in *51<sup>st</sup> AIAA Aerospace Sciences Meeting*, 2013-0774 (Grapevine, TX).
- [7] FIÉVET, R., TINNEY, C.E., MURRAY, N.E., LYONS, G.W. & PANICKAR, P. (2013) "Acoustic Source Indicators using LES in a Fully Expanded and Heated Supersonic Jet," in *19<sup>th</sup> AIAA/CEAS Aeroacoustics Conference*, 2013-2193 (Berlin, Germany).
- [8] BAARS, W.J. & TINNEY, C.E. (2013) "Quantifying Crackle-Inducing Acoustic Shock-Structures Emitted by a Fully-Expanded Mach 3 Jet," in *19<sup>th</sup> AIAA/CEAS Aeroacoustics Conference*, 2013-2081 (Berlin, Germany).
- [9] BAARS, W.J. & TINNEY, C.E. (2013) "A Temporal and Spectral Quantification of the Crackle Component in Supersonic Jet Noise," in *2<sup>nd</sup> Symposium on Fluid-Structure-Sound Interactions and Control*, p. 170 (Shatin, Hong Kong).
- [10] BAARS, W.J., TINNEY, C.E. & WOCHNER, M.S. "Nonlinear Distortion of Acoustic Waveforms from High-Speed Jets," *J. of Fluid Mechanics* [in review].
- [11] BAARS, W.J. (2013) *Acoustics from High-speed Jets with Crackle*, Ph.D. Dissertation, The University of Texas at Austin.
- [12] FIÉVET, R., BAARS, W.J., SILVA, D. & TINNEY, C.E. (2013) "High Fidelity Measurements in the Far-Field of a Mach 3 Jet," in *Bulletin of the American Physical Society*, **58**:18, Abstract D24:7 (Pittsburgh, PA).
- [13] BROCK, B.A., HAYNES, R.H., THUROW, B.S., LYONS, G.W. & MURRAY, N.E. (2014) "An Examination of MHz Rate PIV in a Heated Supersonic Jet," in *AIAA Science and Technology Forum and Exposition (SciTech 2014)*, 2014-1102 (National Harbor, MD).
- [14] BAARS, W.J. & TINNEY, C.E. (2014) "Temporal and Spectral Quantification of the Crackle Component in Supersonic Jet Noise," in Y. Zhou, Y. Liu, L. Huang & D.H. Hodges, eds., *Fluid-Structure-Sound Interactions and Control*, Lecture Notes in Mechanical Engineering, pp. 205-210 (Springer Berlin Heidelberg).

- [15] BAARS, W.J. & TINNEY, C.E. (2014) "POD-Based Spectral Higher-Order Stochastic Estimation," *Physics of Fluids* [in review].
- [16] FIÉVET, R. & TINNEY, C.E. (2014) "Nonlinear Acoustic Distortion in Supersonic Lab-Scaled Jet Vicinity," in *Texas Fluid Dynamics Meeting*.
- [17] FIÉVET, R., TINNEY, C.E., BAARS, W.J. & HAMILTON, M. (2014) "Acoustic Waveforms Produced by a Laboratory Scale Supersonic Jet," *AIAA J.* [in review].

## 5 Planned Activities for Next Reporting Period

The remaining laboratory activities at U. Miss. will be complete during the next reporting period. These measurements will include a survey of the sound field, line-array microphone measurements of the decay in the peak noise direction, LDA measurements of the nozzle exit turbulent boundary layer (relevant for analysis of instability waves of the jet), and additional shadowgraph measurements.



Appendix A: Submitted to *AIAA Journal* for publication:  
**Acoustic Waveforms Produced by a Laboratory Scale Supersonic Jet**  
Fiévet, Timney, Baars & Hamilton



**Acoustic waveforms produced by a laboratory scale  
supersonic jet**

Journal:	<i>AIAA Journal</i>
Manuscript ID:	2014-04-J053538
Manuscript Type:	Full Paper
Date Submitted by the Author:	09-Apr-2014
Complete List of Authors:	Fiévet, Romain; The University of Texas at Austin, Aerospace Engineering and Engineering Mechanics Tinney, Charles; The University of Texas at Austin, Aerospace Engineering and Engineering Mechanics Baars, Woutijn; The University of Melbourne, Mechanical Engineering Hamilton, Mark; The University of Texas at Austin, Mechanical Engineering
Subject Index Category:	20100 Aeroacoustics < 20000 FLUID DYNAMICS, 21700 Supersonic Flow < 20000 FLUID DYNAMICS, 20900 Jets, Wakes, and Viscid-Inviscid Flow Interactions < 20000 FLUID DYNAMICS
Select ONE Subject Index for the Table of Contents.  This is where your paper will show up in the Table of Contents:	20000 FLUID DYNAMICS

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## Acoustic waveforms produced by a laboratory scale supersonic jet.

Romain Fiévet and Charles E. Tinney

*Center for Aeromechanics Research, The University of Texas at Austin, Austin, TX 78712, USA*

Woutijn J. Baars

*The University of Melbourne, Parkville, Victoria 3010, Australia*

Mark F. Hamilton

*Applied Research Laboratories, The University of Texas at Austin, Austin, TX 78713, USA*

The spatial evolution of acoustic waveforms produced by a Mach 3 jet are investigated using both 1/4 inch and 1/8 inch pressure field microphones located along rays emanating from the post potential core where the peak sound emission is found to occur. The measurements are acquired in a fully anechoic chamber where ground, or other large surface reflections are minimal. The calculation of the OASPL along an arc located at 95 jet diameters using 120 planar grid measurements are shown to collapse remarkably well when the arc array is centered on  $x/D_j = 17.5$  which is where the most intense sound is generated. Various statistical metrics, including the quadrature spectral density, number of zero crossings, the skewness of the pressure time derivative and the integral of the negative part of the quadrature spectral density, are exercised along the peak emission path. These metrics are shown to undergo rapid changes within 2 meters from the source regions of this laboratory scale jet. The sensitivity of these findings to both transducer size and humidity effects are discussed. A visual extrapolation of these nonlinear metrics toward the jet shear layer suggests that these waveforms are initially skewed at the source. The agreement between a wave packet modeling of the jet near-field and the experiment allows identifying the transition zone from cylindrical to spherical pressure decay law, and evaluate the source intensity, which is used to estimate the shock formation distance and the Gol'dberg number.

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### Nomenclature

$A$	= non-dimensional wave packet parameter, $L/D_j$
$a_j$	= speed of sound of the jet at the exit of the nozzle, m/s
$a_\infty$	= ambient speed of sound, m/s
$D_j$	= jet exit diameter, m
$f_c$	= cut-off frequency, Hz
$f_j$	= jet characteristic frequency, Hz
$f_0$	= peak spectral frequency along the propagation path, Hz
$G_{pp}$	= one-sided power spectral density of the pressure, $\text{Pa}^2/\text{Hz}$
$L$	= wave packet model scale
$\ell_\alpha$	= absorption length scale
$M_a$	= acoustic Mach number, $U_j/a_\infty$
$M_j$	= gas dynamic Mach number of the jet at the exit of the nozzle, $U_j/a_j$
$M_c$	= Convective Mach number, $U_c/a_\infty$
$N$	= Number of microphones
$N_{zc}$	= Number of zero crossings, $\#/\text{s}$
$p$	= pressure waveform, Pa
$\dot{p}$	= time derivative of the pressure waveform, $\text{Pa}/\text{s}$
$P_0$	= total pressure of the jet at the exit of the nozzle, K
$P_\infty$	= ambient pressure, K
$Q$	= normalized quadrature spectral density, $1/\text{Hz}$
$Q_{neg}$	= integral of the negative part of $Q$
$Q_{p^2p}$	= quadrature spectral density of the pressure, $\text{Pa}^3/\text{Hz}$
$r$	= radial coordinate relative to the jet centerline.
$\bar{r}_c$	= cylindrical shock formation distance, m
$\bar{r}_s$	= spherical shock formation distance, m
$\%RH$	= percent relative humidity
$\gamma$	= skewness factor
$S_{pp}$	= two-sided power spectral density of the pressure, $\text{Pa}^2/\text{Hz}$
$St_{D_j}$	= Strouhal number: $fD_j/U_j$
$\sigma_p$	= standard deviation of $p(t)$
$t$	= time, s
$T_0$	= total temperature of the jet at the exit of the nozzle, K
$T_j$	= static temperature of the jet at the exit of the nozzle, K
$T_\infty$	= ambient temperature, K
$\tau$	= retarded time, s
$U_c$	= convective speed of the jet, m/s
$U_j$	= velocity of the jet at the exit of the nozzle, m/s
$w$	= wave packet amplitude
$W_{error}$	= Wave packet error
$x$	= axial coordinate relative to the jet exit plane
$\bar{x}$	= plane wave shock formation distance
$\alpha$	= absorption factor, $\text{Np}/\text{m}$
$\beta$	= coefficient of nonlinearity
$\Gamma$	= Gol'dberg number for plane waves
$\Lambda_c$	= modified cylindrical Gol'dberg number
$\Lambda_s$	= modified spherical Gol'dberg number
$\phi$	= Mach wave angle measured from the jet centerline, deg.
$\rho$	= distance along a ray emanating from the peak emission location of the jet, m
$\rho_\infty$	= ambient density, $\text{kg}/\text{m}^3$
$\sigma_{p,0}$	= pressure source strength, Pa
$\theta$	= angle measured from the jet centerline, deg.
$\xi_{pp}$	= pressure cross correlation, $\text{Pa}^2$
$FSS$	= Fine Scale Similarity spectra
$LSS$	= Large Scale Similarity spectra
$NPR$	= Nozzle pressure ratio, $P_0/P_\infty$
$OASPL$	= Overall sound pressure level, dB, ref: $20\mu\text{Pa}^2/\text{Hz}$
$PDF$	= Probability density function.

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## I. Introduction

When the convective speed of large scale turbulent structures travel faster than the sound speed of the surrounding gas, they emit Mach waves which propagate at angles relative to the jet axis determined by  $\phi = \cos^{-1}(a_{\infty}/U_c)$  [1-6]. Because these Mach waves are generated by turbulence, they are classified as a component of turbulent mixing noise, which is the most significant source of noise for jets operating at supersonic acoustic Mach numbers. If the amplitude of the Mach wave is large enough to overcome viscous relaxation effects [7, 8], the waveform will steepen with distance from its source, which is more easily observed under full-scale conditions where the characteristic frequencies, and hence viscous relaxation effects, are low [9].

While sawtooth-like waveforms have been observed in both laboratory and full scale jet noise studies, the source and nature of these shapes remains unknown. Most laboratory and full-scale studies are conducted using arrays of microphones placed in the far-field of the jet (beyond 60 or so jet diameters). And so the shape and evolution of these Mach waves, as they propagate from the source regions of the flow to the far-field, are not well understood. A recent study by Baars, Tinney & Wochner (2012) [10] mapped the acoustic field of an unheated supersonic jet (between  $x/D_j = [5 : 10 : 145]$  and  $r/D_j = [25 : 10 : 95]$ ) operating at a gas dynamic Mach number of 3 to unveil the sound field produced by jets comprising supersonic acoustic Mach numbers. Baars & Tinney (2014) [6] later postulated that steepened waveforms are produced either at the source or develop rapidly in the near-field through cumulative nonlinear distortion effects (much earlier than where typical far-field measurements are acquired). Recent Large-Eddy-Simulations by Nichols *et al.* (2013) [11] showed sawtooth waves being emitted from within the jet shear layer and that, despite further steepening from cumulative nonlinear distortions, it was likely that steepened waveforms were also being generated within the jet. Anderson & Freund (2012) [12] observed similar features using direct numerical simulations and went on to illustrate the clumping together of shorter waves in the very near field of supersonic mixing layers. Such observations would suggest that large amplitude waves do not form on their own, but rather form by partnering with other smaller waves to produce larger and more steepened waves further from the jet. Thus, the observations of Nichols *et al.* (2013) [11], Baars & Tinney (2014) [6] and Anderson & Freund (2012) [12] suggest that the

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emulsion of skewed waveforms, from within the jet, proceeded by either rapid waveform steepening or the partnering of similar waves along the propagation path are responsible for sawtooth like structures observed in the far-field. And so, an important question that we seek to address here is whether the acoustic waveforms, produced by Mach waves emanating from laboratory scale jets, undergo cumulative nonlinear distortions, or steepen through other mechanisms.

Here, we will replicate portions of the experiment performed by Baars *et al.* (2012) [10] who used 1/4 inch microphones to characterize the acoustic field of a Mach 3 jet. Contrary to that study, we will employ higher fidelity instruments (1/8 inch pressure field microphones), positioned closer to the jet flow and along lines emanating from the post potential core region where the prominent source of sound is known to reside. As was done before, various well known statistical metrics for quantifying waveform steepening and coalescence will be exercised to determine the shape and nature of the waveforms as they propagate along rays centered on the post-potential core. The effect of absorption on these various metrics will be studied, as the measurements of Baars *et al.* (2012) [10] were conducted at higher relative humidities than the measurements acquired with the 1/8 inch microphones. Doing so will increase the likelihood of capturing, over a broader spectral range, distortions in the acoustic waves as they radiate to the far-field of laboratory scale supersonic jets.

## II. Experimental Description

### A. Facility overview

Both the experiments of Baars *et al.* (2012) [10] and the current higher fidelity measurements were conducted in the fully anechoic chamber and open jet wind tunnel at The University of Texas at Austin. An illustration of the chamber is shown in figure 1 and comprises internal dimensions (wedge-tip to wedge-tip) of 5.74 m(length)×4.52 m(width)×3.66 m(height). The acoustic wedges that are used to produce this echo free environment are composed of 46 cm of fire retardant melamine foam backed by a 10 cm air cavity followed by 14 cm of recycled cotton fibre. The findings from impedance tube tests performed at ETS Lindgren (Cedar Park, Texas) are shown in figure 2 to demonstrate how this particular wedge design produces a normal incidence absorption coefficient of

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99% above 100 Hz for an air cavity depth of 10 cm; the small dip around 250 Hz is due to the flow resistive properties of the melamine foam.

A jet rig is positioned along the centerline of the chamber and tunnel, as shown in figure 1 with a  $1.49 \text{ m}^2$  square aperture on the back wall to allow entrained flow to enter into the chamber. Both the jet and entrained flows exhaust downstream through a  $1.83 \text{ m}^2$  acoustically treated square duct and with the assistance of a vane axial fan to reduce recirculations from forming inside the chamber. The nozzle tested is the same as that described by Baars *et al.* (2012) [10] and was designed to produce a perfectly expanded flow at a gas dynamic Mach number of 3.0 with an exit diameter of  $D_j = 0.0254 \text{ m}$  using unheated air. Nozzle pressure ratios (NPR) are monitored and controlled using a National Instruments CompactRIO system which is capable of reducing variations in the set-point NPR to less than 1% over extended periods of time. Additional details concerning this facility can be found in Donald *et al.* (2014) [13] and Baars (2013) [14].

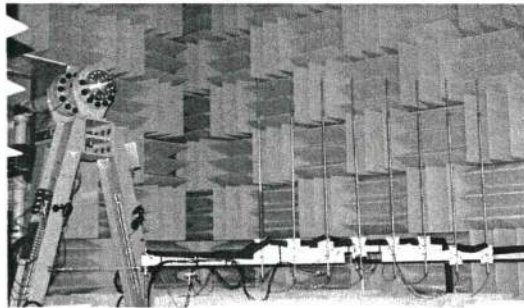


Fig. 1: The fully anechoic chamber at UT-Austin during the current set of measurements.

#### B. Jet operating conditions

Five separate tests were conducted in September 2013 over a duration of 2 days during which atmospheric conditions were carefully recorded; see table 1. While the fully anechoic chamber and wind tunnel are located in a large warehouse; outside air is drawn in to prevent the building from collapsing during runs. Therefore, the environment inside the chamber is that of the air outside of the building. The nozzle was operated at a nozzle pressure ratio of 36.7 which was expected to produce a perfectly expanded flow at the exit of the nozzle (based on quasi 1-D isentropic compressible

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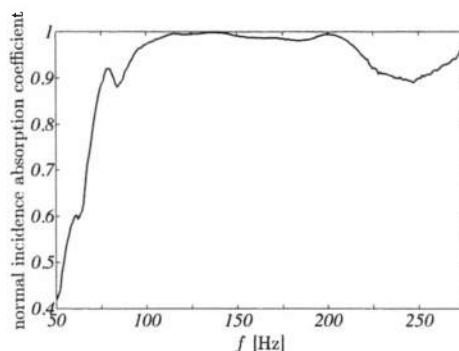


Fig. 2: Normal incidence absorption coefficient of the UT Austin chamber wedge with 10 cm air cavity.

flow equations). Shadowgraph images later revealed the presence of weak shock structures at this pressure ratio. However, no broadband shock tones or screech were observed in any of the acoustic spectra, and so these shocks were not expected to have any influence on the measurements reported here (or those acquired in 2011 with the 1/4 inch microphones [6, 10, 14]). A common set of values for  $P_\infty$ ,  $T_\infty$  and  $T_0$  were chosen by averaging over the test conditions reported in table 1. The resultant differential error between these tests was found to be within 1.5% of the average value. Included in this table are the average values reported by Baars *et al.* (2012) [10], (shown in parenthesis) which compare reasonably well to the conditions reported here. The largest discrepancy between the current set of measurements and the measurements of Baars *et al.* (2012) [10] reside in the atmospheric conditions (relative humidity and atmospheric temperature), with percent relative humidities of 75.5, 63.2 and 48.8 being recorded by Baars *et al.* (2012) [10] on the three days of testing, respectively.

The convective speed of the sound producing events is an important quantity and is assumed to be  $0.8U_j$  [4]. This results in Mach wave angles around  $\phi_{0.8} = 45^\circ$ , which coincide quite well with the peak noise emission angle shown in figure 3; this angle is determined to be where the overall sound pressure level (OASPL) contour, also shown in figure 3 from the  $(x, r)$  grid measurements of Baars *et al.* (2012) [10], is maximum. The lobe-shape pattern in the OASPL is manifest, owing to the competing effects of convection and refraction that is inherent to jet noise. However, closer inspection of the OASPL ridge demonstrates how contour levels, residing closer to the jet, manifest

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Table 1: Testing conditions for the 1/8 inch microphone study. Average values from the 1/4 inch study are shown in parenthesis. Note: the microphone locations corresponding to the  $45_1^\circ$  and  $45_2^\circ$  arrays are provided in table 2.

	Measured values						
Array angle	35°	45 <sub>1</sub>	45 <sub>2</sub>	52°	65°	Average	
$P_\infty$ , kPa	100.0	100.1	100.0	100.2	100.2	100.1 (100.9)	
$P_0/P_\infty$		36.7 ±1% (±4.5%)					36.7
$T_\infty$ , K	302	304	306	303	303	304 (289)	
$T_0$ , K	301	298	301	304	304	302 (288)	
%RH	43	51	40	56	58	N/A	
	Computed values						
$T_j$ , K	108	106	108	109	109	108 (103)	
$a_j$ , m/s	208	207	208	209	209	208 (203)	
$U_j$ , m/s	624	620	624	627	627	624 (609)	
$f_j$ , kHz	24.6	24.4	24.6	24.7	24.7	24.6 (24.0)	
$M_j = U_j/a_j$	3.0	3.0	3.0	3.0	3.0	3.0	
$M_a = U_j/a_\infty$	1.78	1.78	1.78	1.80	1.80	1.79 (1.79)	
$M_c = 0.8U_j/a_\infty$	1.43	1.42	1.42	1.44	1.44	1.43 (1.43)	
$\phi_{0.8}$ , deg.	45	45	45	46	46	45 (46)	
$\phi_{0.7}$ , deg.	37	36	37	37	37	37 (37)	

Mach waves with angles ( $\phi$ ) coinciding with convective speeds on the order of  $0.7U_j$  (identified in figure 3 by  $\phi_{0.7} = 35^\circ$ ). Contrarily, in regions further from the jet, Mach wave angles coincide with structures moving at  $0.8U_j$  (identified in figure 3 by  $\phi_{0.8} = 45^\circ$ ). Variations in the Mach wave ridge are attributed to differences in the convective speed of the more pronounced sound producing events and their locations in the post potential core region relative to one another.

### C. Instrumentation and microphone placement

Because results from two different studies employing different kinds of instruments are being evaluated, the details concerning both are provided here for completeness. The first of the acoustic measurements is described by Baars *et al.* (2012)[10] and was performed in December of 2011 using four 1/4 inch prepolarized, pressure-field condenser microphones (PCB model 377B10 capsules with 426B03 preamplifiers) positioned at grazing incidence to the jet axis. These IEPE type transducers were powered using a NI PXI-4472 card which then digitized the signals at 102.4 kS/s with 24 bit resolution ( $\pm 10\text{v}$ ) for a minimum of  $2^{20}$  samples. Signal attenuation occurs at 0.84 times the Nyquist frequency with the built in anti-aliasing filters on the NI PXI-4472 card. The placement of these microphones involved both a 2-D grid of points in the  $(x, r)$  plane as well as a line array along

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the peak emission path, which was found to emanate from a region centered around  $x/D_j = 17.5$ . Spectral based metrics are constructed using  $2^{13}$  samples per block, thus resulting in a frequency resolution of  $\delta f = 12.5$  Hz.

As for the current set of measurements, eight 1/8 inch pressure-field microphones (G.R.A.S. Type 46DD capsules with G.R.A.S. Type 26CB preamplifiers) were placed at grazing incidence, as shown in figure 1. These microphones are capable of resolving signals over a broader range of frequencies (from 6.5 Hz up to 140 kHz with  $\pm 2$  dB error) with a dynamic range from 46 dB to 179 dB. IEPE power was supplied by a Dytran Instruments Inc. 4121 power supply, which were then low pass filtered using a Kliron-Hite corp filter (model FLX-3007). These low pass filters employ a 4<sup>th</sup> order Butterworth filter and were custom designed to have a cut off frequency of 140 kHz. Microphone signals were digitized simultaneously at 500 kS/s per channel using two NI PXI-6122 boards (four single ended channels per board with 16 bit resolution over  $\pm 5$ v) for 13.1 seconds uninterrupted. This largely over-sampled rate ensures better resolution in the rise time of the shock-structures, which is a crucial point in this study. One sided power spectral densities were then generated from 400 blocks of data, each comprising  $2^{14}$  samples per block and a frequency resolution of  $\delta f = 30.52$  Hz.

The 1/8 inch microphone measurements were conducted using four different rays centered on  $x/D_j = 17.5$ , as is shown in figure 3. For high Mach number supersonic jets, the region in the flow between the collapse of the potential core and the supersonic core length are known to be the location where turbulent mixing is not only most intense, but also occurs at supersonic velocities [15, 16]. Identifying the correct propagation path is not necessarily trivial as the jet does not behave like a single compact source. This is especially problematic close to the hydrodynamic region where pressure waves comprise a superposition of both acoustic and evanescent signatures [17]. Choosing the wrong path artificially triggers the apparition of nonlinear distortions in the signal that one might interpret as a sign of wave steepening; this problem has been addressed in past studies [10, 18]. Here we attempted to place the microphones as close to the jet as possible without them being endangered, and without contaminating the signals with evanescent signatures. The growth of the jet shear layer was approximated to be  $0.1x$  based on the measurements of Tinney *et al.* (2008) [19].

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Table 2: Locations of 1/8 inch microphones.

$\theta$	offset from source	$\Delta\rho$
$65^\circ$	$13.0D_j$	$10D_j$
$52^\circ$	$13.0D_j$	$10D_j$
$45_1^\circ$	$13.0D_j$	$5D_j$
$45_2^\circ$	$35.5D_j$	$10D_j$
$35^\circ$	$43.5D_j$	$10D_j$

A more conservative estimate of  $0.15x$  was employed as is shown in figure 3. Table 2 identifies the location of the 1/8 inch microphone rays and is listed in terms of the ray angle (measured relative to the jet centerline), the starting distance of each ray relative to the post potential core region, as well as the separation distance between subsequent microphones on each ray.

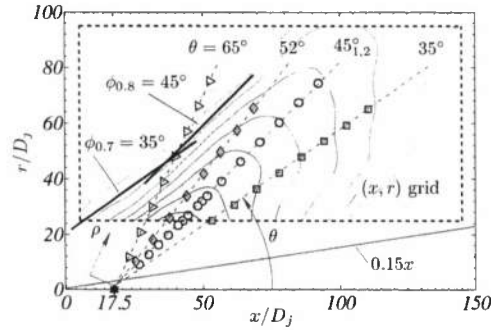


Fig. 3: Location of 1/8 inch microphone rays and coordinate system. Contour of OASPL (dB, ref:  $20\mu\text{Pa}^2/\text{Hz}$ ) from 1/4 inch  $(x, r)$  grid measurements of Baars & Tinney (2014).

In an effort to remove any possible contaminations produced by low frequency reflections in the chamber, a second order high-pass filter was applied to the data with a cut-off frequency of 300 Hz. Prior to this, corrections based on manufacturer supplied data, was utilized to correct for the frequency dependent error in each transducer. As is shown in figure 4a, the effect of this frequency correction on the spectral shape is subtle, but was used nonetheless. Additional studies were performed with and without the microphone's protective caps, as well as without the low-pass antialiasing filter. No discernible trends were observed. And so, both the protective caps and low-pass filters were utilized for the remainder of the studies.

A sample illustration of the final spectra is shown in figure 4b, with comparison to the FSS and LSS spectra of Tam *et al.* (1996) [20]. The two microphones are selected from the  $35^\circ$  and  $65^\circ$  rays

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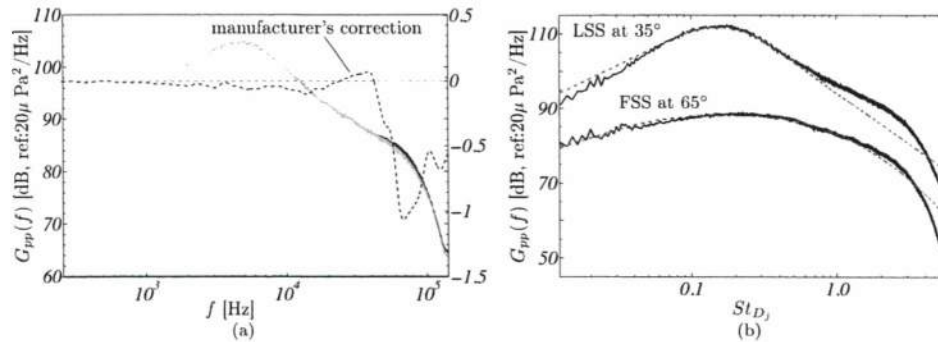


Fig. 4: (a) One sided power spectral density (PSD) of a sample microphone signal at  $45^\circ$  with and without corrections using manufacturer supplied data. (b) PSD at  $\theta = 35^\circ$  and  $65^\circ$  being compared to the LSS and FSS of Tam *et al.* (1996).

and are located at  $85.3D_j$  and  $83.0D_j$  from the jet source. The spectra corresponding to the  $65^\circ$  observer is located outside of the Mach cone, and so it is characterized by a broader profile and a smooth roll-off, whereas the shallow angle observer at  $35^\circ$  comprises a noticeable characteristic peak with relatively linear decay. The good agreement between the experimental data and the similarity spectra is encouraging and demonstrates the universality of the spectra produced by high Mach number jet flows.

### III. Sound pressure decay in the jet far-field

One of the underlying challenges in implementing the model developed by Baars (2013) [14], which predicts shock formation distances and the degree of cumulative nonlinear distortion in the acoustic waveform from high speed jets, is that it requires one to know the size and amplitude of the source. This is a difficult proposition, even with the advent of more sophisticated numerical simulations at our disposal; consider the different source definitions floating about the open literature [21–23]. The experimental handicap is pretty straightforward: simply placing a pressure sensing element closer to a jet flow manifests a flow-biased description of the source properties, given the blending of both acoustic and hydrodynamic waves within the jet's hydrodynamic periphery. Two established methods, one described by Tinney & Jordan (2008) [17], the other by Grissi & Camussi (2012) [24], offer some relief in separating these two components of the pressure field, but

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are limited to flows comprising sound producing events with convective speeds that are, on average, subsonic. And so, an adequate means by which the source mechanisms, responsible for generating Mach waves, can be isolated and studied remains to be found.

A resurgence in recent years with the use of wave packets (Morris (2009) [25], Papamoschou (2011) [26], Kuo *et al.* (2012) [27] and Jordan & Colonius (2013) [28]) may offer some relief, and at a reduced cost. Modeling the evolution of supersonically convecting large-scale instabilities by wave packets provides a theoretical framework for studying the noise radiated by supersonic jets. In figure 5 (taken from Appendix A of Baars [14]), the standard deviation ( $\sigma_p$ ) of the radiating component of the pressure signal along the Mach wave radiation angle, and emanating from the source region, is shown to encompass three distinct zones: one that abides by cylindrical decay ( $\sigma_p \propto \rho^{-0.5}$ ), another that abides by spherical decay ( $\sigma_p \propto \rho^{-1.0}$ ) and a third intermediate zone that transitions the two. Certainly the location where this transition occurs is quite important in determining source amplitude. Baars (2013) [14] shows how the answers to this can be expressed as a ratio of the wave packet model scale ( $L$ ) relative to the jet diameter, such that  $A = L/D_j$ ; the effect of  $A$  on the three decay regions is shown in figure 5 for a source located on the lip line of the nozzle. Thus, to determine the proper values associated with both the wave packet scale and its location ( $r_{wp}$ ), a sensitivity study was conducted. For this, the spatial limits of the Gaussian envelop for all cases was confined to three standard of deviations while the wave packet amplitude was adjusted using pressure levels measured halfway along the array. Wave packet parameters  $r_{wp}$  and  $A$  that were finally chosen were based on a least error sum between all available microphones and the wave packet. The error function expression for this is defined as follows,

$$W_{error} = \sum_{i=1}^N \frac{[(w_i - m_i)(N - i)]^2}{m_i^2}. \quad (1)$$

where  $w_i$  corresponds to the wave packet pressure value located at the  $m_i$  microphone and  $i$  increases with increasing distance from the source. Furthermore, measurements closer to the jet were weighted heavier (using  $N - i$ ) than measurements further away, given the larger changes in pressure amplitude that occur closer to the source region. The results of this sensitivity study are shown in figure 6

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where the location of the source is expressed by its position along the propagation path ( $\rho_0$ ). Additional lines have been added which identify the final wave packet parameters ( $A = 8.75$  and  $\rho_0 = (\sqrt{2}/2)D_j$ ) that were selected for the remainder of the analysis, as well as the location of the jet shear layer. Thus, the large-scale instability waves are assumed to initiate their development at the exit plane of the nozzle and grow along the nozzle lip line to their peak level at  $x/D_j = 17.5$  before decaying to negligible levels far downstream. Under these assumptions,  $2L = 17.5$ . In figure 7, the measured pressure decay (using both the 1/4 inch and 1/8 inch microphones placed along the  $45^\circ$  path) compares favorably well to the wave packet model. Deviations from the cylindrical and spherical decay laws are found to be within 2% of the wave packet model at  $\rho/D_j = 12.3$  and 33.4, respectively. The findings are a testament to both the scale of the wave packet chosen, as well as the propagation path along which the microphones are placed.

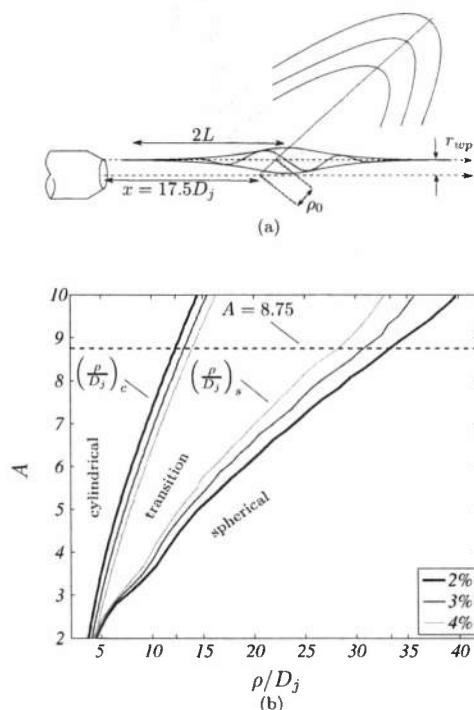


Fig. 5: (a) schematic of the wave packet scale versus jet scale. (b) Regions where the wave packet model obeys cylindrical and spherical spreading along the peak noise path. Contour lines define error thresholds of 2%, 3% and 4%.

Figures 8a and 8b present the decay and inverse decay, respectively, measured using the 1/8 inch

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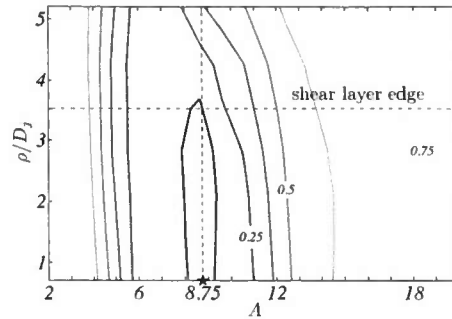


Fig. 6: Integrated errors between the wave packet model and laboratory measurements for various wave packet length scales and source locations along the propagation path. Contour levels have been normalized so that they range between 0 and 1 with increments of 0.125.

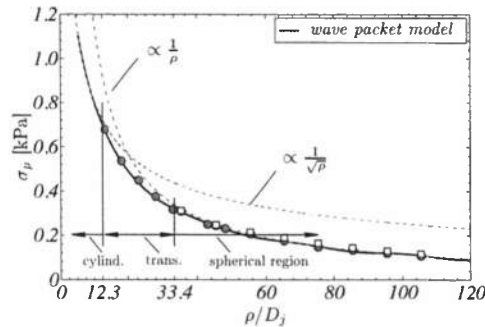


Fig. 7: Spreading trends along the peak noise emission path obtained from the wave packet model using  $A = 8.75$ . Both experimental datasets are shown with  $\square$  corresponding to the 1/4 inch microphones.

microphones along all four rays. The agreement between the  $45^\circ$  and  $52^\circ$  rays is quite remarkable and confirms how linear decay holds well over a range of angles centered on the Mach wave angle. Thus one should expect some flexibility in both the convective speed of the relevant sound producing events, as well as the acoustic region where the propagation path represents that of a wave-guide so to speak; spherical decay in the acoustic pressure is not restricted to a single angle. Albeit, the  $45^\circ$  ray is the only arrangement where a higher spatial resolution permits a verification that this trend holds closer to the jet. Not so surprisingly is the fact that the rate of decay at steeper ( $65^\circ$ ) and shallower ( $35^\circ$ ) angles is much greater; the  $65^\circ$  array being much more extreme. Glancing at the OASPL contours in figure 3 justifies why this is expected.

A further observation is made with the trends presented in figures 7 and 8. Foremost, the

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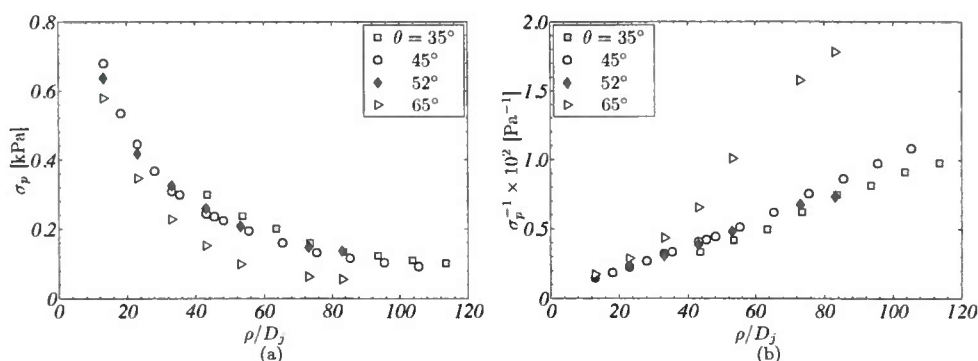


Fig. 8: (a)  $\sigma_p$  decay and (b)  $\sigma_p^{-1} \times 10^2$  from 1/8 inch microphones.

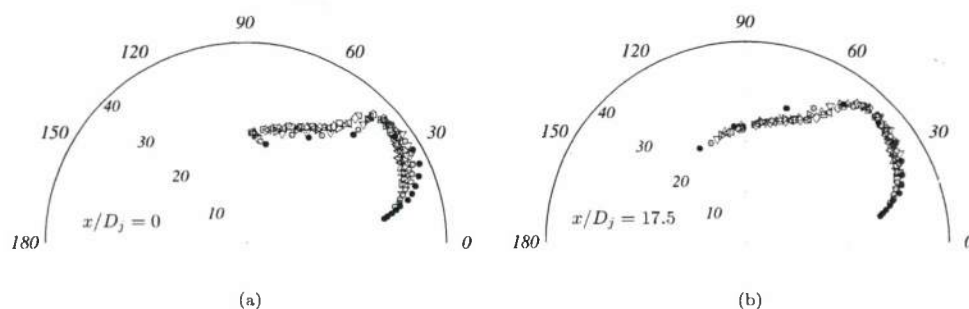


Fig. 9: OASPL (dB, ref:  $20\mu\text{Pa}^2/\text{Hz}$ ) from  $(x, r)$  plane grid measurements of Baars & Tinney (2014) projected to an arc array at  $\rho/D_j = 95$  using spherical decay. Assuming source region at (a)  $x/D_j = 0$  and (b)  $x/D_j = 17.5$ . The values shown are less 100dB.

remarkable agreement between the measurements acquired with the 1/4 inch and 1/8 inch microphones demonstrates how transducer size and atmospheric absorption have little effect on the pressure variance (expressed here as OASPL). And so, the topography of the OASPL in figure 3 using 1/4 inch microphones is considered an accurate mapping of the pressure intensity in the far-field of this Mach 3 jet flow. Likewise, because the sound pressure is shown to decay spherically after approximately 35 diameters, any prediction of the far-field pressure using measurements outside of this location, and assuming spherical decay, should provide an accurate mapping of the OASPL in the far-field, so long as the correct propagation path is employed. It is important that we emphasize how the location in which the acoustic pressure abides by spherical decay is based on the operating conditions of the nozzle and is expected to scale with acoustic Mach number. Most often

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the propagation path is chosen to emanate from the jet exit plane. Baars & Tinney (2014) [6] shows how various features in the far-field of supersonic jets collapse exceptionally well when the prominent source region, located after the collapse of the potential core, is chosen. A demonstration of this is shown in figure 9 where the OASPL from the entire  $(x, r)$  grid measurements of Baars & Tinney (2014) [6] (120 measurement points in total) are projected to an arc array situated at  $\rho/D_j = 95$  using spherical decay. In figure 9a the source is assumed to reside at the nozzle exit plane at  $x/D_j = 0$ , while in figure 9b the source is moved downstream to  $x/D_j = 17.5$ . The findings are remarkable and expose the kinds of gross errors that will occur when one chooses to propagate jet noise data along rays emanating from the jet exit plane; nearly identical observations have been made in the rocket noise community [15, 29]. The only discrepancies in figure 9b are with the closer measurement points (identified as filled and open circles) that were located at  $\rho/D_j < 35$ , and is anticipated since these waveforms will initially undergo cylindrical decay (as shown in figure 7), which was not corrected for in this simple exercise. Thus, both fine-scale similarity and large-scale similarity like regions in the far-field can be predicted from measurements closer to the jet itself, so long as the correct propagation path is employed.

#### IV. Analysis of acoustic waveforms along the peak emission path

Baars & Tinney (2014) [6] demonstrated how the statistical properties of the acoustic waveforms are relatively unchanged along rays emanating from the prominent source regions of supersonic jets; similar results were shown here in section III. Furthermore, only weak waveform distortions were measured using 1/4 inch microphones placed in the mid to far-field regions of the jet and under higher levels of humidity. And so, the motivation of this part of our study is twofold. The first is to demonstrate the effect of sensor size and atmospheric absorption on capturing these distortions, while the second is to address the postulations of Baars & Tinney (2014) [6] through a thorough analysis of the data acquired with the higher fidelity instruments.

The statistical metrics that are employed comprise the quadrature spectral density, skewness of the pressure time derivative, the number of zero crossings per unit time ( $N_{zc}$ ) and the integral of the negative component of the quadrature spectral density. These metrics have been thoroughly

described elsewhere and are used to determine the degree of nonlinearity in the acoustic waveform at a single point. When analyzed over several points along the propagation path, the changes to the waveform amount to nonlinearities encumbered from cumulative distortions. This is true so long as the measured waveforms are from a single source, or have limited interference from neighboring wavefronts produced by other nearby sources in the flow. The latter will give rise to artificial steepening and the formation of rogue waves formed from coalescence. Other metrics were considered and included the wave steepening factor and Kurtosis, though neither revealed anything significant and so they are not shown here. A listing of these metrics is provided as follows:

- **Quadrature Spectral Density.** The definition for this metric is well documented in the work of Morfey & Howell (1981) [30] and Howell & Morfey (1987) [31]. This quantity highlights the importance of non-linear distortion across the spectrum relative to the power spectral density; the normalized forms of this indicator is as follows,

$$Q = \frac{Q_{p^2p}}{\sigma_p^3}, \quad (2)$$

$$\frac{Q}{S} = \frac{Q_{p^2p}/\sigma_p^3}{S_{pp}/\sigma_p^2} \quad (3)$$

where  $Q_{p^2p} = -\Im(2p_2(f)p^*(f))$ , and  $p_2(f)$  and  $p(f)$  are the Fourier transforms of the signals  $p(t)$  and  $p^2(t)$ , respectively. The denominator in Eq. (3) comprises the ensemble averaged power spectral density, which is simply  $S_{pp} = \langle p(f)p^*(f) \rangle$ . Howell & Morfey (1987) [31] recognized early on the limitation in using these metrics at a single point and that the problem [of predicting distortion] remains unsolved without estimating  $Q_{p^2p}$  as a function of distance. This is especially problematic with experimental data, where transducer errors inherently prevent the signal from being truly Gaussian; thus it is difficult to envision this term ever being zero in a laboratory environment. Figure 8a illustrates how the pressure standard deviation along the 45° ray abides by realistic decay laws, and so an application of both Eqs. (2) and (3) to this data should provide meaningful results.

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- **Skewness of the pressure derivative.** The amount of wave steepening, and thus the degree of nonlinearity in the pressure signal, can be quantified by computing the probability density function (PDF) and skewness factor of the pressure time derivative  $\dot{p}(t)$ . The PDF is denoted as  $B(\dot{p})$  while its asymmetry is defined by the third central moment,

$$\overline{\dot{p}^3} \equiv \int_{-\infty}^{\infty} \dot{p}^3 B(\dot{p}) d\dot{p}. \quad (4)$$

The dimensionless asymmetry is thus the well known skewness factor, and is defined as

$$\gamma \equiv \frac{\overline{\dot{p}^3}}{\sigma_p^3}, \quad (5)$$

where  $\sigma_p^2$  is the variance of  $\dot{p}(t)$ . Numerous applications using this metric to quantify nonlinearities in real waveforms can be found in the open literature with the skewness of the pressure time derivative providing the most sensitivity [32].

- **Number of zero crossings per unit time.** Since higher amplitude waves move faster along the propagation path and merge or coalesce with weaker waveforms, the number of zero crossings will decrease with increasing distance from the source. This does not necessarily mean that the waveform is steepening along the propagation path and so the link between  $N_{zc}$  and cumulative nonlinear distortion effects warrants caution.
- **Integral of negative  $Q$ .** If cumulative distortions are occurring along the propagation path, then one would anticipate a flux of energy in and out of the mid frequencies. This can be quantified by integrating the negative part of the quadrature spectral density at each location on the ray as follows,

$$Q_{neg}(\rho) = \int_{Q < 0} Q(f, \rho) df. \quad (6)$$

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### A. Spectral indicators

Profiles of  $Q$  and  $Q/S$  are shown in figures 10a and 10b using equations (2) and (3), respectively, along the four different rays (a 1% bandwidth moving filtered has been employed). As expected, both  $Q$  and  $Q/S$  profiles reflect a loss in energy at mid frequencies, with  $Q/S$  demonstrating an increase in energy at high frequencies. Closer inspection reveals two distinct trends. The first comprises grey lines with arrows indicating the direction of increasing distance from the source and with noticeable changes in both energy and peak frequency. Increases in mid-frequency energy suggest waveform coalescence while increases in high frequency energy suggests waveform steepening. The other set, highlighted by solid dark lines, encompasses regions that reside beyond approximately  $40D_j$ . These black lines collapse reasonably well and suggest that the acoustic waveforms are no longer coalescing or distorting along the propagation path; this agrees well with the measurements of Baars & Tinney (2014) [6] using the 1/4 inch microphones. Along the  $65^\circ$  ray, the trends are difficult to decipher. Since this ray is outside the Mach cone, the lack of any discernible trend is expected. Albeit, the near perfect collapse of the OASPL shown in figure 9b is a testament to the choice of propagation path, even for the sideline and steep angle observer. This is consistent with the measurements of Baars & Tinney (2014) [6] where it has been shown that shock structures are concentrated mainly along the Mach wave angle.

### B. Scalar indicators

Figure 11 compares the number of zero crossings per second, the skewness of the pressure derivative and the integral of  $Q_{neg}$  along the  $45^\circ$  path. The data acquired with the 1/8 inch microphones are analyzed using both the original sample rate of 500 kHz as well as a reduced rate of 100 kHz, which coincides with the sample rate used by the 1/4 inch microphones. The findings from the latter are used to compare the 1/8 inch microphone measurements with the data acquired using the 1/4 inch microphones in an effort to determine the effect of transducer size and atmospheric properties (humidity and temperature) on these higher order metrics. For each metric, an arbitrary approximation of the trend corresponding to the 500 kHz sampling of the 1/8 inch microphone is shown, along with its spatial derivative. These derivatives reveal three distinct zones. The first

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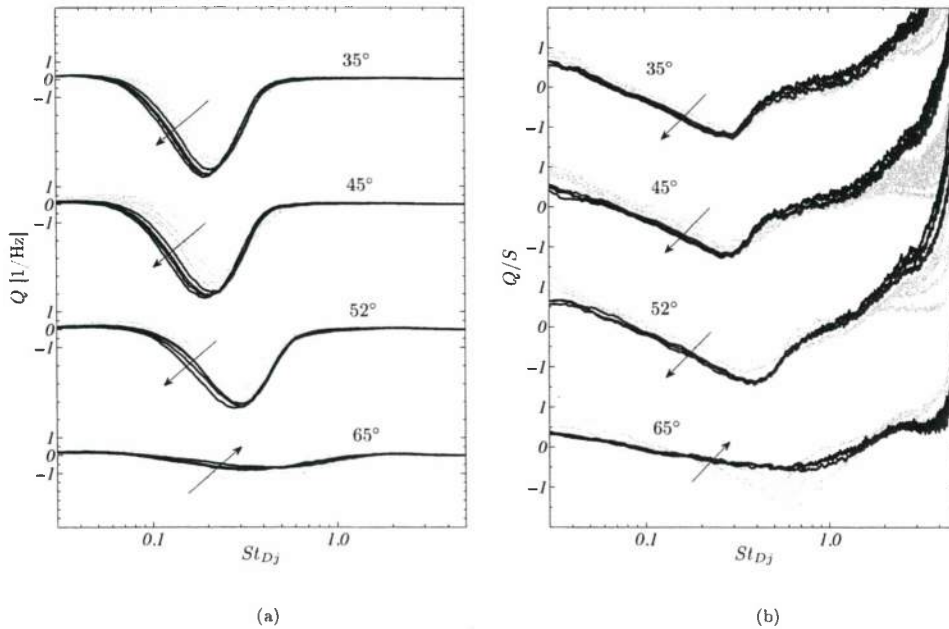


Fig. 10: (a)  $Q \times 10^{-5}$  and (b)  $Q/S$ , Morfey-Howell indicators (1% bandwidth moving filter) for all microphones (gray) and for the last 4 microphones (black).

manifests a fairly constant spatial rate of change while the second reveals changes in amplitude along the propagation path. The third zone identifies a region where the metric is saturated and is no longer changing.

Where the number of zero crossings and the skewness of the pressure time derivative are concerned, the effect of sampling rate is clearly manifest. However the effects of temperature and humidity on the number of zero crossing and the integral of  $Q_{neg}$  appear not to be affected. While corrections for atmospheric absorption are well established, they cannot be corrected in the raw time signal. And so the effects of atmospheric absorption can only be ascertained from raw measurements of this kind.

As for the skewness of the pressure derivative  $\gamma(\dot{p})$ , several relevant features are seen in figure 11b. First of all, three trends corresponding to each data-set (same legend as in figure 11a) are clearly present. The new data-set sampled at 500 kHz shows a quasi-linear growth of  $\gamma(\dot{p})$  that suggests wave steepening; zero skewness is reflective of a normal distribution. By 80 or so jet di-

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ameters ( $\rho = 2$  m) the steepening process appears to have reached its maximum at which point the trend remains either constant or begins to decay. For the resampled data (1/8 inch microphones resampled at 100 kHz), the change in  $\gamma(\dot{p})$  follows the originally sampled data consistently up until roughly 0.7 meters from the source region, after which changes in the skewness begin to reflect the trends produced by the 1/4 inch microphone study. This agreement demonstrates how distortion mechanisms affect the low frequency bands in the waveform earlier on as the lower sampled data reflects that of the higher sampled data. Differences between the resampled 1/8 inch microphone data and the 1/4 inch microphone measurements are attributed to elevated levels of humidity during the 1/4 inch study. A visual extrapolation of the trends toward the origin ( $\rho \rightarrow 0$ ) suggests that the waveforms produced in the source region of the jet are already skewed, thus confirming the simulations of Nichols *et al.*(2013) [11].

Figure 11c quantifies the net flux of energy transfer across the quadrature spectral density shown in figure 10a. By restricting the limits of integration to only the negative part of the spectra, which is shown to be concentrated on the mid frequency range, any noise inherent to the borders of the spectrum are avoided. This registers the flux of mid-range spectral energy that is lost to waveform coalescence and is therefore a relevant indicator. Once again, a trend very similar to  $\gamma(\dot{p})$  appears with a noticeably linear growth at the start of the array. Further away the evolution starts to decrease and eventually converges as the spatial rate of change becomes negligible.

The evolution of the spatial rate-of-change along the array for these three indicators are shown alongside one another in figure 12 with each one being normalized by its maximum amplitude in order to compare trends. Overall, these indicators comprise a similar shape, which would suggest that both waveform steepening and coalescence are occurring within 2.5 meters or so from the source region of this laboratory scale jet.

### C. Shock formation distance

In an effort to link the observed phenomena to the theoretical estimates for shock formation distance, two such formulations are considered, one for spherically propagating waves and the other for cylindrical waves. Given the changes in pressure amplitude registered by our measurements

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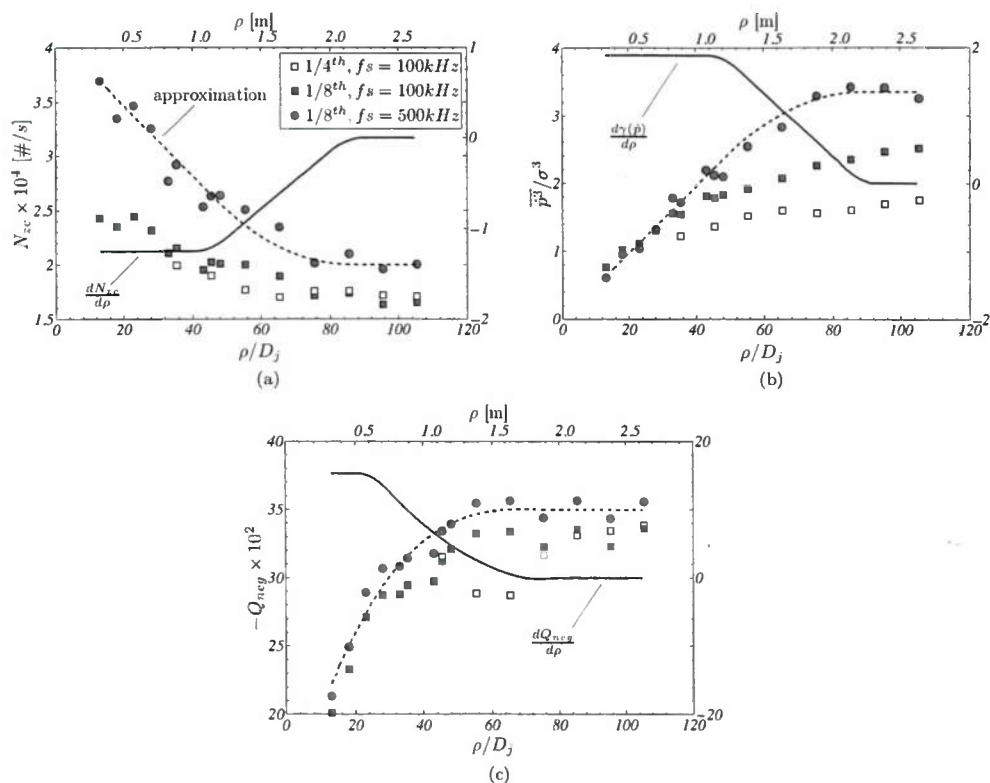


Fig. 11: (a) Number of zero-crossings, (b) skewness of the pressure time derivative and (c)  $-Q_{neg} \times 10^2$  along the  $45^\circ$  ray.

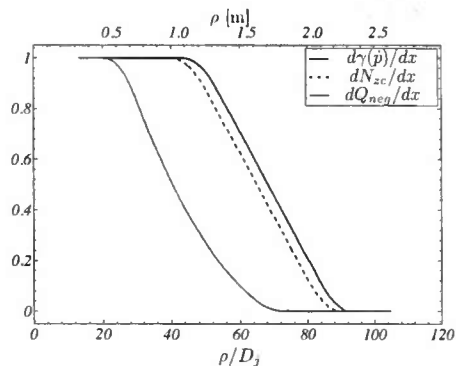


Fig. 12: Comparison of the spatial derivative of  $N_{zc}$ ,  $\gamma(p)$  and  $Q_{neg}$  normalized by their respective maximum value.

and confirmed by wave packet modeling, the real solution will be a blended estimate of the two.

Following Hamilton & Blackstock (2008) [8], the shock formation distance for an initially broadband

gaussian waveform that propagates cylindrically and spherically from its source in a lossless fluid can be estimated from, respectively,

$$\bar{r}_c = r_0(1 + \bar{x}/2r_0)^2 \quad (7)$$

and

$$\bar{r}_s = r_0 \exp(\bar{x}/r_0), \quad (8)$$

where  $\bar{x}$  is the plane wave shock formation distance defined by Gurbatov & Rudenko (p.383 of Hamilton & Blackstock [8]) as,

$$\bar{x} = \frac{\rho_\infty a_\infty^3}{\beta(2\pi f_0)\sigma_{p,0}}, \quad (9)$$

and  $r_0$  is the source radius. The coefficient of nonlinearity is given by  $\beta = (\gamma + 1)/2 = 1.2$  while the density of air is determined from table 1 and ideal gas properties to be  $\rho_\infty = 1.14 \text{ kg/m}^3$ . Figure 4b reveals a characteristic frequency valued at  $f_0 = 3.6 \text{ kHz}$  for the  $45^\circ$  ray, which is the same used by Baars (2013) [14]. Both the size and strength of the source are not easily determined, given the experimental and numerical handicaps discussed in section III. Likewise, expressions for shock formation distance require that  $kr \gg 1$ , that is, the source radius must exceed the maximum wavelength of the radiated sound. Three likely locations for the source surface are considered in this exercise, though an infinite number of solutions can be obtained. The first borders the periphery of the jet shear layer at  $\rho = 3.5D_j$  by assuming a shear layer growth of  $0.1x$  [19]. The second is selected to replicate the location used by Baars (2013) [14] so that  $r_0$  is located at  $\rho = 2.5D_j$ . The third location is assumed to be located on the nozzle lip-line at  $\rho = (\sqrt{2}/2)D_j$ , where the peak noise source has been shown to reside [35]. Each of these locations is within the cylindrical decay region of the pressure amplitude, and so the source amplitudes (for these three points) are extrapolated using the wave packet solution for  $L = 8.75$  and assuming  $p \propto 1/\sqrt{\rho}$ . Shock formation distances are then estimated using Eqs. (7) and (8) for cylindrical and spherical decay along the entire propagation

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Table 3: Shock formation distances and effective Gol'dberg numbers.

$r_0, D_j$	0.7	2.5	3.5	2.5(Baars 2013)
$kr_0$	1.2	4.1	5.8	4.2
$\sigma_{p,0}, \text{Pa}$	2900	1546	1300	4948
$\bar{x}, \text{m}$	0.62	1.17	1.39	0.36
$\bar{r}_c, \text{m}$	6.04	6.60	6.85	0.92
$\bar{r}_s, \text{m}$	$10^{13}$	$10^7$	$10^5$	17.2
$\Gamma$	618	329	277	1171
$\Lambda_c$	22.9	21.3	21.1	217
$\Lambda_s$	$10^{-21}$	$10^{-10}$	$10^{-8}$	0.18

path, respectively. Thus, for a starting waveform that abides by gaussian statistics and is allowed to propagate in the absence of viscous absorption effects, its shock formation distance will be bounded by  $\bar{r}_c$  and  $\bar{r}_s$ .

Findings from this analysis are shown in table 3 alongside the original estimates from Baars (2013) [14] who estimated the source amplitude to be a factor of three larger than what is measured here; Baars (2013) [14] assumed the source amplitude abided by spherical propagation along its entire path, which is reasonable, considering the information that was available at the time. Nonetheless, the recipe outlined by Baars (2013) [14] is employed here and provides reasonable estimates for  $\bar{r}_c$  and  $\bar{r}_s$ . The findings suggest that, aside from both the effects imposed by having presteepened waveforms at the source as well as viscous relaxation along the propagation path, it is unlikely that shocks will form within the confines of this laboratory facility.

#### D. Gol'dberg number

Naturally, the next step is to calculate the Gol'dberg number, a measure of the strength of nonlinear distortion relative to that of dissipation. When  $\Gamma < 1$ , attenuation dominates and the formation of shocks is suppressed. When  $\Gamma \gg 1$  nonlinear effects will be distinctly present. The expression for plane waves is well established and is defined as,

$$\Gamma = \frac{\ell_\alpha}{x} \quad (10)$$

where the absorption length scale ( $\ell_\alpha$ ) is simply the inverse of the absorption coefficient ( $\alpha$ ) which is evaluated in table 3 using an expression from Bass *et. al.* (1996) [36]. We evaluate  $\alpha$  using the

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characteristic frequency  $f_0 = 3.6$  kHz and for a relative humidity determined by the average value recorded during the  $45^\circ_{1,2}$  tests (%RH= 45). The resultant absorption for the 1/8 inch microphone study is found to be  $\alpha = 0.0026$  Np/m, which yielded  $\ell_\alpha = 385$  m ( $\ell_\alpha = 417$  m for the 1/4 inch microphone study conducted by Baars *et al.* (2012) [10] based on 70% relative humidity and an atmospheric temperature of 290 K). Surprisingly, the differences in absorption between the current measurements and the 1/4 inch study is small; changes in temperature are balanced by changes in humidity for this particular frequency. As was done for the shock formation distance, an expression for the Gol'dberg number for cylindrically and spherically propagating waves is needed. Hamilton (2013) [37] recently showed how the effective Gol'dberg number for diverging waves can be expressed by a single parameter as follows,

$$\Lambda_c = \frac{\Gamma}{1 + \pi\bar{x}/4r_0} \quad (11)$$

$$\Lambda_s = \Gamma \exp(-\pi\bar{x}/2r_0) \quad (12)$$

where Eq. (11) and (12) are the cylindrical and spherical forms of the expression. Estimates of  $\Lambda_c$  and  $\Lambda_s$  for the 1/8 inch and 1/4 inch studies are provided in table 3 with contradictory conclusions being offered. That is, estimates that assume purely cylindrical decay suggest cumulative nonlinear distortions will be present in the waveform, whereas estimates assuming pure spherical decay suggest that no such distortions will form.

#### E. Direct observation

A comparison of the instantaneous waveforms registered along the  $45^\circ$  ray in the inner and outer zones are shown in figures 14 and 15, respectively. These waveforms are aligned by cross-correlating the first microphone signal with subsequent signals on the array, where  $\xi_{pp'}(\tau) = \langle p(\rho, t)p(\rho', t) \rangle$ . The corrected time delay is identified by the maximum correlation, as shown in figure 13, which exposes two important features. That is, correlations of this magnitude demonstrate how directive



the waveforms are along this path, while the decay in correlation level with increasing distance is evidence of contamination from other waveforms produced by other sources located elsewhere in the flow. And so, this raises additional concerns as to whether cumulative waveform distortions are forming beyond the hydrodynamic near-field regions of laboratory scale jets, aside from the statistical analysis provided in Section IV A.

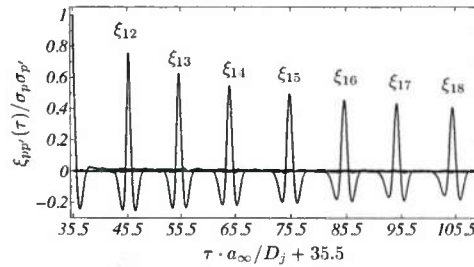


Fig. 13: Cross correlations between microphones on the  $45^\circ$  ray in the outer zone encompassing  $\rho/D_j = [35.5 : 10 : 105.5]$ .

Turning our attention to the raw pressure time series in figures 14 and 15, several features are manifest, which complement the simulations of Nichols *et al.* (2013) [11] and Anderson & Freund (2012) [12]. Albeit, without knowing the particle velocity simultaneously, the discussion and interpretation provided here warrants caution. In figures 14 and 15, several additional lines have been drawn with labels corresponding to the following kinds of events:

- Wave Steepening (WS). This corresponds to instances when the wave registered at multiple locations on the ray appears to be the same wavefront and is steepening. This is identified by a single wave that evolves from a more relaxed wave shape to a N-wave shape with increasing distance.
- Wave Passing (WP). These waves correspond to instances where the direction of propagation differs slightly from the  $45^\circ$  path along which the microphones are placed. Because the width of the wavefront is finite, it either appears or disappears in the microphone signal at various points consecutively. It is assumed that the number of waves entering and exiting the propagation path is the same.
- Rogue Wave (RW). These will form when two smaller waveforms travel at small angles relative

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to one another and eventually merge to produce a much larger and steepened wave whose amplitude is approximately that of the original waves combined.

- Wave Coalescence (WC). This would appear similar to a rogue wave where two waves, propagating along the same path, merge and then coalesce.

The significance of having both passing waves and rogue waves occurring along the propagation path is that both will artificially escalate  $N_{zc}$ ,  $\gamma(\dot{p})$  and  $Q_{neg}$  and mislead one into believing that cumulative distortions are occurring along the propagation path. For the sample time series shown in figures 14 and 15, instances of passing waves is significant. Rogue waves only appear in these time series at further distances in figure 15 where N-wave like structures are also significant.

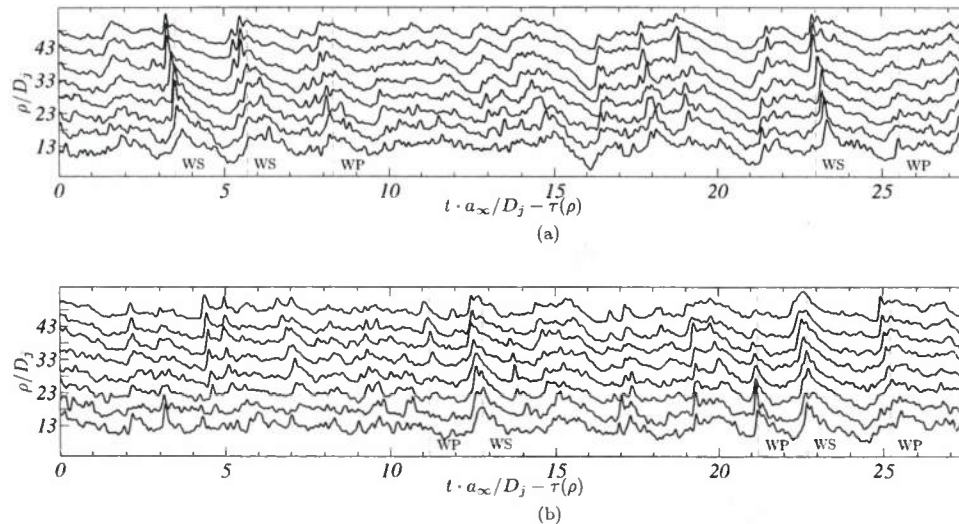


Fig. 14: Evolution of the raw pressure time series (shown as  $p(\rho, t)/\sigma_p(\rho)$ ) along the  $45^\circ$  ray in the inner zone encompassing  $\rho/D_j = [13 : 5 : 48]$ .

## V. Conclusions

The acoustic waveforms produced by an unheated Mach 3 jet are investigated using 1/4 inch microphones covering a 2-D grid in the  $(x, r)$  plane as well as 1/8 inch microphones placed along rays emanating from the source regions of the flow at  $x/D_j = 17.5$ . Various metrics including pressure amplitude decay, the normalized quadrature spectral density, skewness of the pressure derivative,

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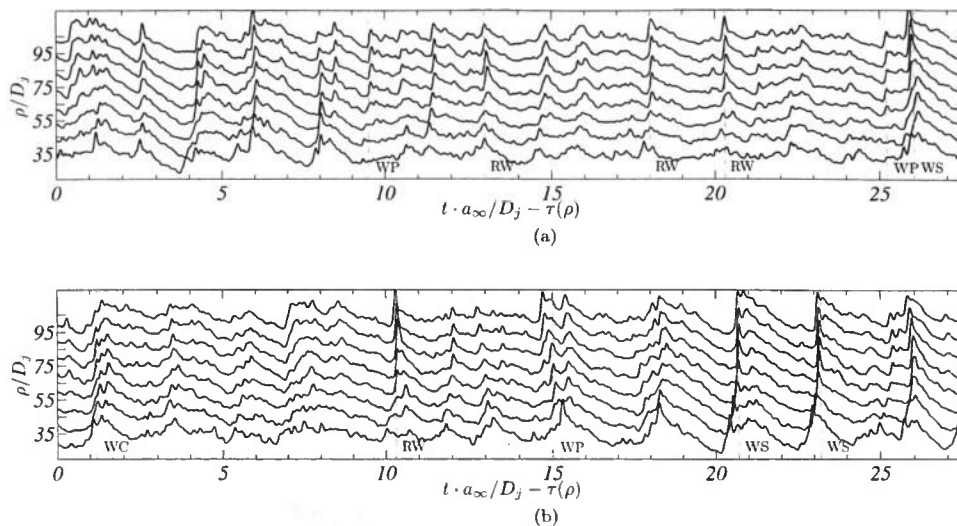


Fig. 15: Evolution of the raw pressure time series (shown as  $p(\rho, t)/\sigma_p(\rho)$ ) along the  $45^\circ$  ray in the outer zone encompassing  $\rho/D_j = [35.5 : 10 : 105.5]$ .

the number of zero crossings and the integral of the negative part of the quadrature spectral density were used to study the behavior of these waveforms along the propagation path. Where the pressure amplitude is concerned, the location where its rate of decay first abides by cylindrical spreading, followed by spherical spreading, was identified and was confirmed using a simple wave packet model. The OASPL appeared not to be affected by sensor size and so the linear prediction of its amplitude at 95 jet diameters was performed and shown to collapse remarkably well for all azimuthal angles when propagated along rays emanating from the post potential core regions of the jet.

Both spectral and scalar metrics used to compute nonlinear waveform distortions revealed three different areas in the jet near-field. The first comprised a zone where waveform distortions appear to be increasing at a constant rate that initiate in close proximity to the jet (within one meter from the prominent source region). A second zone showed a decrease in nonlinear distortion effects while a third zone demonstrated how the waveforms were no longer steepen and had either reached their maximum or were beginning to undergo a state of relaxation. A visual extrapolation of the nonlinear metrics towards the source region suggests that the acoustic waveforms are initially distorted at the source. Consequently, a direct observation of the raw acoustic time series along the propagation path revealed both passing waves and rogue waves in the pressure waveform, both of which would

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ultimately escalate the metrics for quantifying cumulative nonlinear distortions.

## VI. Acknowledgments

This work is funded through the Office of Naval Research Jet Noise Reduction program, contract number N00014-11-1-0752, under the direction of Drs. Joseph Doychak and Brenda Henderson.

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Appendix B: 2014 Texas Fluid Dynamics Meeting Presentation  
**Nonlinear Acoustic Distortion in Supersonic Lab-Scaled Jet Vicinity**  
Fiévet & Timney

Texas Fluid Dynamics Meeting, 2014

**Nonlinear acoustic distortion in supersonic lab-scaled jet vicinity.****R. Fiévet & C. E. Tinney***The University of Texas at Austin**Aerospace Engineering & Engineering Mechanics, Austin, Texas, 78712.**email: fievetromain@gmail.com, cetinney@utexas.edu**website: <http://www.ae.utexas.edu/facultysites/tinney/>*

The spatial evolution of acoustic waveforms produced by a Mach 3 jet are investigated in a fully anechoic chamber using both 1/4 inch and 1/8 inch pressure field microphones located along rays emanating from the post potential core where the peak sound emission is found to occur. Predictions of the OASPL along an arc located at 95 jet diameters using 120 planar grid measurements are shown to collapse remarkably well when the arc array is centered on  $x/D_j = 17.5$  (post potential core), as opposed to the jet exit plan. Statistical metrics such as the quadrature spectral density, number of zero crossings, the skewness of the pressure time derivative and the integral of the negative part of the quadrature spectral density are used. The results demonstrate how distorted waves form through coalescence and wave steepening and occur within a few meters from the source in these laboratory scale jets. The remarkable agreement between a wavepacket modeling of the jet near-field and the experiment allows identifying the transition zone from cylindrical to spherical pressure decay law, and demonstrates how the latter of the two occurs within 35  $D_j$  along the propagation path.