

W.A. Good

331 commonly used sections

AD-A280 301



0

NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS

DTIC
ELFOTE
MAY 21 1994
S F D

REPORT No. 315

AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI

By
NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS

94-15791



419



94 5 25 078

DTIC QUALITY INSPECTED 3

This document has been approved
for public release and sale; its
distribution is unlimited.

~~RESTRICTED ONLY~~

UNITED STATES
GOVERNMENT PRINTING OFFICE
WASHINGTON : 1929

AERONAUTICAL SYMBOLS

1. FUNDAMENTAL AND DERIVED UNITS

	Symbol	Metric		English	
		Unit	Symbol	Unit	Symbol
Length.....	<i>l</i>	meter.....	m	foot (or mile).....	ft. (or mi.)
Time.....	<i>t</i>	second.....	sec	second (or hour).....	sec. (or hr.)
Force.....	<i>F</i>	weight of one kilogram.....	kg	weight of one pound.....	lb.
Power.....	<i>P</i>	kg/m/sec.....		horsepower.....	HP.
Speed.....		km/hr.....		mi./hr.....	M. P. H.
		m/sec.....		ft./sec.....	f. p. s.

2. GENERAL SYMBOLS, ETC.

W, Weight, = mg

g, Standard acceleration of gravity = 9.80665 m/sec.² = 32.1740 ft./sec.²

m, Mass, = $\frac{W}{g}$

ρ , Density (mass per unit volume).

Standard density of dry air, 0.12497 (kg·m⁻⁴ sec.²) at 15° C and 760 mm = 0.002378 (lb.-ft.⁻⁴ sec.²).

Specific weight of "standard" air, 1.2255 kg/m³ = 0.07651 lb./ft.³

mk^2 , Moment of inertia (indicate axis of the radius of gyration, *k*, by proper subscript).

S, Area.

S_w, Wing area, etc.

G, Gap.

b, Span.

c, Chord length.

b/c, Aspect ratio.

f, Distance from *c. g.* to elevator hinge.

μ , Coefficient of viscosity.

3. AERODYNAMICAL SYMBOLS

V, True air speed.

q, Dynamic (or impact) pressure = $\frac{1}{2} \rho V^2$

L, Lift, absolute coefficient $C_L = \frac{L}{qS}$

D, Drag, absolute coefficient $C_D = \frac{D}{qS}$

C, Cross-wind force, absolute coefficient

$$C_C = \frac{C}{qS}$$

R, Resultant force. (Note that these coefficients are twice as large as the old coefficients *L_C*, *D_C*.)

i_w, Angle of setting of wings (relative to thrust line).

i_t, Angle of stabilizer setting with reference to thrust line.

γ , Dihedral angle.

$\frac{Vl}{\mu}$, Reynolds Number, where *l* is a linear dimension.

e. g., for a model airfoil 3 in. chord, 100 mi./hr. normal pressure, 0° C: 255,000 and at 15° C., 230,000;

or for a model of 10 cm chord 40 m/sec, corresponding numbers are 299,000 and 270,000.

C_p, Center of pressure coefficient (ratio of distance of *C. P.* from leading edge to chord length).

β , Angle of stabilizer setting with reference to lower wing, = (*i_t* - *i_w*).

α , Angle of attack.

ϵ , Angle of downwash.

REPORT No. 315

AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI
CONTINUATION OF REPORTS Nos. 93, 124, 182, 244, and 286

By
**NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS**

39331-29-1

381

Accession For	
NTIS CRA&I	<input checked="" type="checkbox"/>
DTIC TAB	<input checked="" type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By _____	
Distribution /	
Availability Codes	
Dist	Avail and/or Special
A-1	
12	

DTIC QUALITY INSPECTED 3

~~"DTIC USE ONLY"~~

TABLE OF CONTENTS

	Page
Introduction.....	385
Transformation constants.....	385
Chart index.....	387
Index of abbreviations.....	387
Group index.....	388
Alphabetical index.....	389
Airfoil sections.....	390-414
United States sections.....	390-392
German sections.....	392-399
French sections.....	399-404
Belgium sections.....	505-407
Italian sections.....	407-414
Charts Nos. 21, 22, 23, 24.....	415-418

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

NAVY BUILDING, WASHINGTON, D. C.

(An independent Government establishment, created by act of Congress approved March 3, 1915, for the supervision and direction of the scientific study of the problems of flight. It consists of 12 members who are appointed by the President, all of whom serve as such without compensation.)

JOSEPH S. AMES, Ph. D., *Chairman*.
Provost, Johns Hopkins University, Baltimore, Md.
DAVID W. TAYLOR, D. Eng., *Vice Chairman*.
Washington, D. C.
CHARLES G. ABBOT, Sc. D.,
Secretary, Smithsonian Institution, Washington, D. C.
GEORGE K. BURGESS, Sc. D.,
Director, Bureau of Standards, Washington, D. C.
WILLIAM F. DURAND, Ph. D.,
Professor Emeritus of Mechanical Engineering, Stanford University, California.
JAMES E. FECHET, Major General, United States Army,
Chief of Air Corps, War Department, Washington, D. C.
WILLIAM E. GILLMORE, Brigadier General, United States Army,
Chief, Matériel Division, Air Corps, Wright Field, Dayton, Ohio.
EMORY S. LAND, Captain, United States Navy,
Assistant Chief, Bureau of Aeronautics, Navy Department, Washington, D. C.
CHARLES F. MARVIN, M. E.,
Chief, United States Weather Bureau, Washington, D. C.
WILLIAM A. MOFFETT, Rear Admiral, United States Navy,
Chief, Bureau of Aeronautics, Navy Department, Washington, D. C.
S. W. STRATTON, Sc. D.,
President Massachusetts Institute of Technology, Cambridge, Mass.
ORVILLE WRIGHT, Sc. D.,
Dayton, Ohio.

GEORGE W. LEWIS, *Director of Aeronautical Research*.
JOHN F. VICTORY, *Secretary*.
HENRY J. E. REID, *Engineer in Charge, Langley Memorial Aeronautical Laboratory,*
Langley Field, Va.
JOHN J. IDE, *Technical Assistant in Europe, Paris, France.*

EXECUTIVE COMMITTEE

JOSEPH S. AMES, *Chairman*.
DAVID W. TAYLOR, *Vice Chairman*.

CHARLES G. ABBOT.
GEORGE K. BURGESS.
JAMES E. FECHET.
WILLIAM E. GILLMORE.
EMORY S. LAND.

CHARLES F. MARTIN.
WILLIAM A. MOFFETT.
S. W. STRATTON.
ORVILLE WRIGHT.

JOHN F. VICTORY, *Secretary*.

REPORT No. 315

AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI

CONTINUATION OF REPORTS NOS. 93, 124, 182, 244, AND 286

By NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

INTRODUCTION

This collection of data on airfoils has been made from the published reports of a number of the leading aerodynamic laboratories of this country and Europe.¹ The information which was originally expressed according to the different customs of the several laboratories is here presented in a uniform series of charts and tables suitable for the use of designing engineers and for purposes of general reference.

It is a well-known fact that the results obtained in different laboratories, because of their individual methods of testing, are not strictly comparable even if proper scale corrections for size of model and speed of test are supplied. It is, therefore, unwise to compare too closely the coefficients of two wing sections tested in different laboratories. Tests of different wing sections from the same source, however, may be relied on to give true relative values.

The absolute system of coefficients has been used, since it is thought by the National Advisory Committee for Aeronautics that this system is the one most suited for international use and yet it is one from which a desired transformation can be easily made. For this purpose a set of transformation constants is given.

Each airfoil section is given a reference number, and the test data are presented in the form of curves from which the coefficients can be read with sufficient accuracy for designing purposes. The dimensions of the profile of each section are given at various stations along the chord in per cent of the chord length, the latter also serving as the datum line. The shape of the section is also shown with reasonable accuracy in order to enable one to more clearly visualize the section under consideration, the outside of the heavy line representing the profile.

The authority for the results here presented is given as the name of the laboratory at which the experiments were conducted, as explained under abbreviations, with the size of model, wind velocity, and year of test.

TRANSFORMATION CONSTANTS

For the convenience of those who prefer to use a system of units other than the absolute system, there is given below a table of transformation constants based on the standard condition adopted by the National Advisory Committee for Aeronautics of—

Temperature	= 15° C	= 59° F.
Pressure	= 760 mm Hg	= 29.92 in. Hg.
Humidity	= 0	
Gravity	= 9.80665 m/s ²	= 32.1740 ft./sec. ²

thus giving values of specific weight of air

$$W = 1.2255 \text{ kg/m}^3 = 0.07651 \text{ lb./cu. ft.}$$

¹ A previous collection of airfoil sections numbered 1 to 759 and Charts 1 to 20 may be found in N. A. C. A. Reports Nos. 93, 124, 182, 244, and 286.

and of density

$\rho = 0.12497 \text{ kg-m}^{-4}\text{s}^2$ in the French engineering or kilogram, meter, second system.

Or

$\rho = 0.002378 \text{ lb.-ft.}^{-4}\text{sec.}^2$ in the English or pound, foot, second system.

In absolute units	$P = CV^2\rho/2$
In kg/m^2 (m/s)	$P = .0625 CV^2$
In kg/m^2 (km/h)	$P = .004822 CV^2$
In lb./sq. ft. (ft./sec.)	$P = .001189 CV^2$
In lb./sq. ft. (mi./hr.)	$P = .002558 CV^2$

(Note that these constants are half as large as those used in Reports Nos. 93 and 124 and that the absolute coefficients used in this report are twice as large as the old coefficients. See Report No. 240 regarding change in absolute coefficients.)

INDEX

Four separate types of indexes are given—chart indexes which make it possible for a designer to select the wing section most suitable for the particular design in which he is interested; a group index which is arranged by countries and laboratories at which tests were conducted, each section also being designated by a reference number; an index of abbreviations, used on the characteristic sheets, to indicate the laboratories at which the tests were made; and an alphabetical index.

CHART INDEX

In order that the designer may easily pick out a wing section which is suited to the type of airplane on which he is working, four index charts are given which classify the wings according to their aerodynamic and structural properties. In the charts of this report a lower-case letter is placed adjacent to the reference number giving Vl values, so that a comparison can be made without referring to the individual drawings. In this value V represents the wind velocity in feet per second and l a linear dimension, the chord length in feet.

In chart No. 21 the minimum drag C_D , is plotted against the L/D at one-fourth the maximum lift C_L . This chart should be used in choosing a wing section for a high-speed airplane, the wing sections being more suited for this use the farther they are from the lower left-hand corner.

In chart No. 22 the mean spar depth is plotted against the maximum lift C_L , in order to show the possible strength and lightness of the wing structure. The higher the maximum lift coefficient is, the smaller will be the wing area and the lighter the structural weight, and in the same way the greater the depth of the spars the lighter will be their weight, so that the sections the greatest distance from the lower left-hand corner will give the lightest and strongest wings. The "mean spar depth" is obtained by assuming the spars to be located, respectively, at 15 and 60 per cent of the chord, and by dividing the sum of their thicknesses, in per cent of chord length at these points, by 2.

In chart No. 23 the maximum L/D is plotted against the maximum lift C_L , which is of use in choosing the wing section for a slow and efficient airplane. In the same way as before the sections farthest from the lower left-hand corner are the best for this purpose.

In chart No. 24 the L/D at two-thirds the maximum lift C_L , is plotted against the maximum lift C_L . This chart can be used for choosing a section that will give an efficient climb or a long range at cruising speed. The best sections for this purpose will be farthest from the lower left-hand corner of the chart.

CHART INDEX

	Page
Chart No. 21. Minimum drag C_D , plotted against L/D at one-fourth the maximum lift C_L	415
Chart No. 22. Mean spar depth plotted against the maximum lift C_L	416
Chart No. 23. Maximum L/D plotted against maximum lift C_L	417
Chart No. 24. L/D at two-thirds the maximum lift C_L , plotted against the maximum lift C_L	418

INDEX OF ABBREVIATIONS

Name of laboratory at which tests were made	Abbreviations used on figures
Langley Memorial Aeronautical Laboratory of the National Advisory Committee for Aeronautics, U. S. A.	L. M. A. L.
Washington Navy Yard, U. S. A.	W. N. Y.
Engineering Division, McCook Field, U. S. A.	McC. F.
Ergebnisse der Aerodynamischen Versuchsanstalt zu Göttingen, Germany	Göttingen.
Service Technique de l'Aéronautique, France	S. T. Aé.
Laboratoire Aerotechnique de Rhode St. Genese-Bruxelles, Belgium	Rhode St. Genese.
Instituto Sperimentale Aeronautico, Italy	I. S. A.

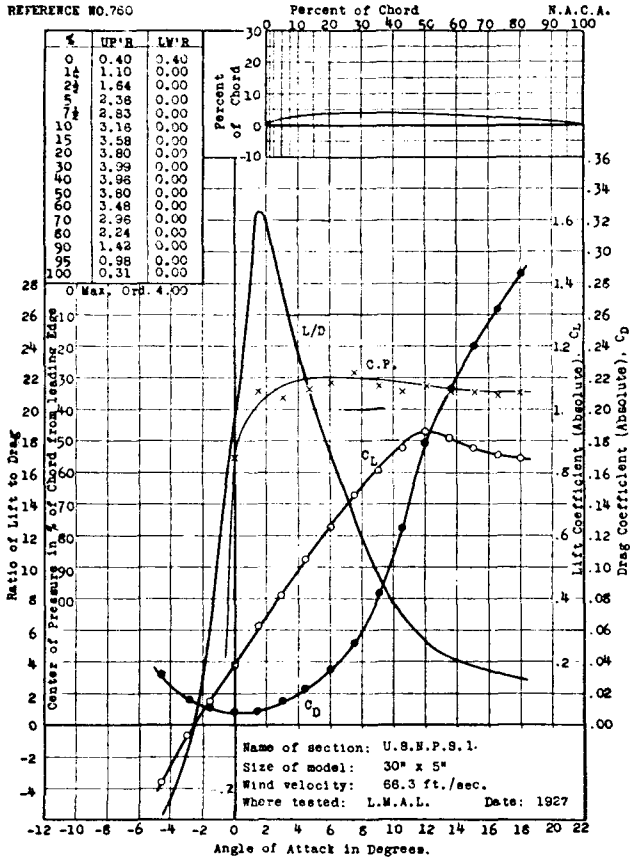
GROUP INDEX

Airfoil	Wind tunnel where tested	Report reference number	Airfoil	Wind tunnel where tested	Report reference number
UNITED STATES			FRANCE—continued		
U. S. N. P. S. 1	L. M. A. L.	760	St. Cyr 162 (Royer)	S. T. Aé. Lab.	810
U. S. N. P. S. 2	do.	761	St. Cyr 170 (Royer)	do.	811
U. S. N. P. S. 3	do.	762	St. Cyr 174 (Royer)	do.	812
U. S. N. P. S. 4	do.	763	St. Cyr 233 (Bartel 37-IIb)	do.	813
U. S. N. P. S. 5	do.	764	St. Cyr 235 (Bartel 37-Ic)	do.	814
U. S. N. P. S. 6	do.	765	St. Cyr 237 (Bartel 15-Ic)	do.	815
Martin M-I	W. N. Y.	766	St. Cyr 239 (Bartel 37-IIa)	do.	816
Martin M-II	do.	767	St. Cyr 240 (Bartel 37-IIb)	do.	817
Martin M-III	do.	768	St. Cyr 242 (Bartel 57-IIc)	do.	818
R-3 (Root section)	McC. F.	769	St. Cyr 243 (Bartel 37-IIIc)	do.	819
U. S. A. 45M	do.	770			
GERMANY			BELGIUM		
Göttingen 359	Göttingen	771	Rhode St. Genese 1	Rhode St. Genese	820
Göttingen 361	do.	772	Rhode St. Genese 2	do.	821
Göttingen 362	do.	773	Rhode St. Genese 16	do.	822
Göttingen 368	do.	774	Rhode St. Genese 17	do.	823
Göttingen 369	do.	775	Rhode St. Genese 19	do.	824
Göttingen 371	do.	776	Rhode St. Genese 26	do.	825
Göttingen 372	do.	777	Rhode St. Genese 29	do.	826
Göttingen 373	do.	778	Rhode St. Genese 31	do.	827
Göttingen 374	do.	779	Rhode St. Genese 33	do.	828
Göttingen 375	do.	780	Rhode St. Genese 35	do.	829
Göttingen 377	do.	781	Rhode St. Genese 37	do.	830
Göttingen 392	do.	782			
Göttingen 397	do.	783	ITALY		
Göttingen 399	do.	784	I. S. A. 334	I. S. A.	831
Göttingen 401	do.	785	I. S. A. 390	do.	832
Göttingen 402	do.	786	I. S. A. 472	do.	833
Göttingen 403	do.	787	I. S. A. 500	do.	834
Göttingen 408	do.	788	I. S. A. 501	do.	835
Göttingen 417	do.	789	I. S. A. 502	do.	836
Göttingen 428	do.	790	I. S. A. 507	do.	837
Göttingen 437	do.	791	I. S. A. 605	do.	838
Göttingen 438	do.	792	I. S. A. 663	do.	839
Göttingen 439	do.	793	I. S. A. 693	do.	840
Göttingen 442	do.	794	I. S. A. 695	do.	841
Göttingen 443	do.	795	I. S. A. 768	do.	842
Göttingen 444	do.	796	I. S. A. 776	do.	843
Göttingen 445	do.	797	I. S. A. 801	do.	844
			I. S. A. 803	do.	845
			I. S. A. 804	do.	846
			I. S. A. 805	do.	847
			I. S. A. 806	do.	848
			I. S. A. 807	do.	849
			I. S. A. 809	do.	850
			I. S. A. 812	do.	851
			I. S. A. 820	do.	852
			I. S. A. 821	do.	853
			I. S. A. 829	do.	854
			I. S. A. 858	do.	855
			I. S. A. 881a	do.	856
			I. S. A. 881b	do.	857
			I. S. A. 881c	do.	858
			I. S. A. 911	do.	859
FRANCE					
Eiffel 359 (Nieuport Astra)	S. T. Aé. Lab.	798			
Eiffel 386 (S. T. Ae.)	do.	799			
Eiffel 387 (S. T. Aé.)	do.	800			
Eiffel 402 (Pescara)	do.	801			
Eiffel 429 (Lachassagne)	do.	802			
Eiffel 432 (Lachassagne)	do.	803			
Eiffel 433a (Chalambel)	do.	804			
Eiffel 438 (Lachassagne)	do.	805			
St. Cyr 152 (Royer)	do.	806			
St. Cyr 153 (Royer)	do.	807			
St. Cyr 156 (Royer)	do.	808			
St. Cyr 157 (Royer)	do.	809			

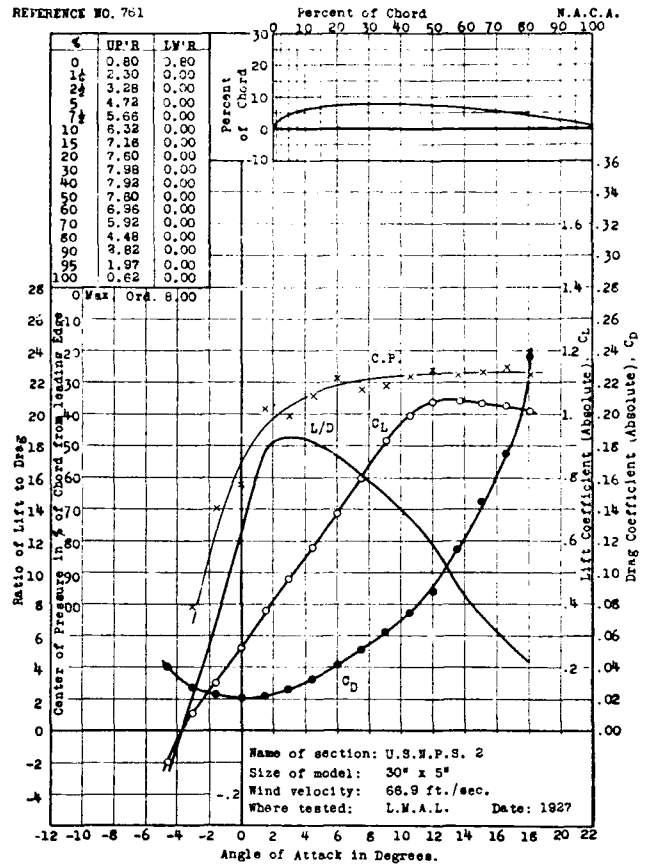
ALPHABETICAL INDEX

Airfoil	Report reference number	Airfoil	Report reference number
Eiffel 359 (Nieuport Astra)	798	I. S. A. 804	846
Eiffel 386 (S. T. Ac.)	799	I. S. A. 805	847
Eiffel 387 (S. T. Ac.)	800	I. S. A. 806	848
Eiffel 402 (Pescara)	801	I. S. A. 807	849
Eiffel 429 (Lachassagne)	802	I. S. A. 809	850
Eiffel 432 (Lachassagne)	803	I. S. A. 812	851
Eiffel 433a (Chalambel)	804	I. S. A. 820	852
Eiffel 438 (Lachassagne)	805	I. S. A. 821	853
Göttingen 359	771	I. S. A. 829	854
Göttingen 361	772	I. S. A. 858	855
Göttingen 362	773	I. S. A. 881a	856
Göttingen 368	774	I. S. A. 881b	857
Göttingen 369	775	I. S. A. 881c	858
Göttingen 371	776	I. S. A. 911	859
Göttingen 372	777	Martin M-I	766
Göttingen 373	778	Martin M-II	767
Göttingen 374	779	Martin M-III	768
Göttingen 375	780	R-3 (Root Section)	769
Göttingen 377	781	Rhode St. Genese 1	820
Göttingen 392	782	Rhode St. Genese 2	821
Göttingen 397	783	Rhode St. Genese 16	822
Göttingen 399	784	Rhode St. Genese 17	823
Göttingen 401	785	Rhode St. Genese 19	824
Göttingen 402	786	Rhode St. Genese 26	825
Göttingen 403	787	Rhode St. Genese 29	826
Göttingen 408	788	Rhode St. Genese 31	827
Göttingen 417	789	Rhode St. Genese 33	828
Göttingen 428	790	Rhode St. Genese 35	829
Göttingen 437	791	Rhode St. Genese 37	830
Göttingen 438	792	St. Cyr 152 (Royer)	806
Göttingen 439	793	St. Cyr 153 (Royer)	807
Göttingen 442	794	St. Cyr 156 (Royer)	808
Göttingen 443	795	St. Cyr 157 (Royer)	809
Göttingen 444	796	St. Cyr 162 (Royer)	810
Göttingen 445	797	St. Cyr 170 (Royer)	811
I. S. A. 334	831	St. Cyr 174 (Royer)	812
I. S. A. 390	832	St. Cyr 233 (Bartel 7-Ib)	813
I. S. A. 472	833	St. Cyr 235 (Bartel 37-Ic)	814
I. S. A. 500	834	St. Cyr 237 (Bartel 15-Ic)	815
I. S. A. 501	835	St. Cyr 239 (Bartel 37-IIa)	816
I. S. A. 502	836	St. Cyr 240 (Bartel 37-IIb)	817
I. S. A. 507	837	St. Cyr 242 (Bartel 57-IIc)	818
I. S. A. 605	838	St. Cyr 243 (Bartel 37-IIIc)	819
I. S. A. 663	839	U. S. A. 45M	770
I. S. A. 693	840	U. S. N. P. S. 1	760
I. S. A. 695	841	U. S. N. P. S. 2	761
I. S. A. 768	842	U. S. N. P. S. 3	762
I. S. A. 776	843	U. S. N. P. S. 4	763
I. S. A. 801	844	U. S. N. P. S. 5	764
I. S. A. 803	845	U. S. N. P. S. 6	765

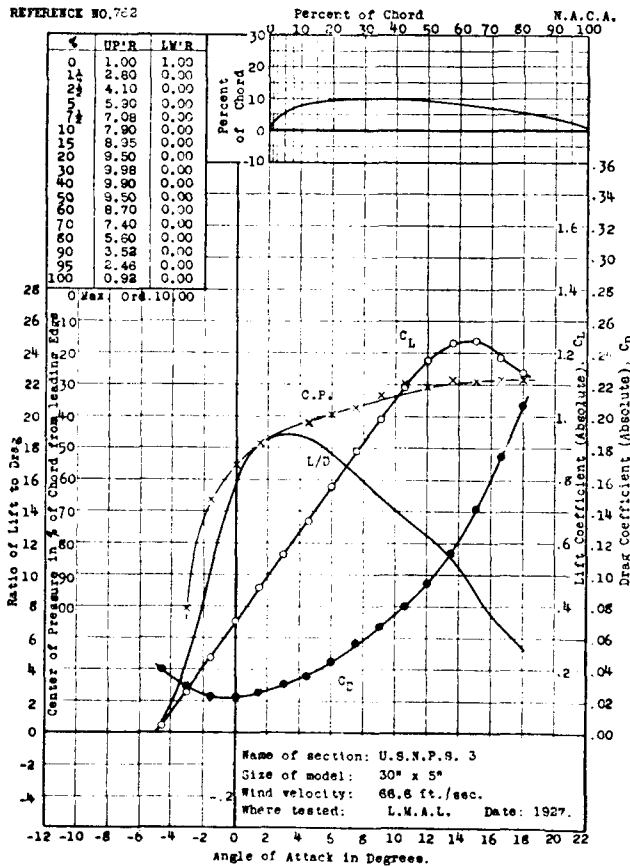
REFERENCE NO. 760



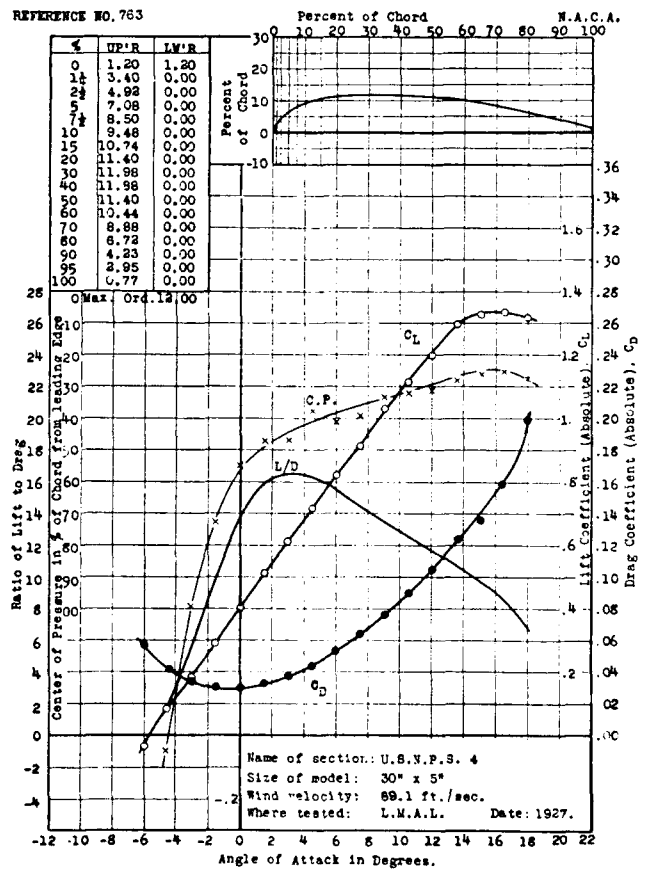
REFERENCE NO. 761



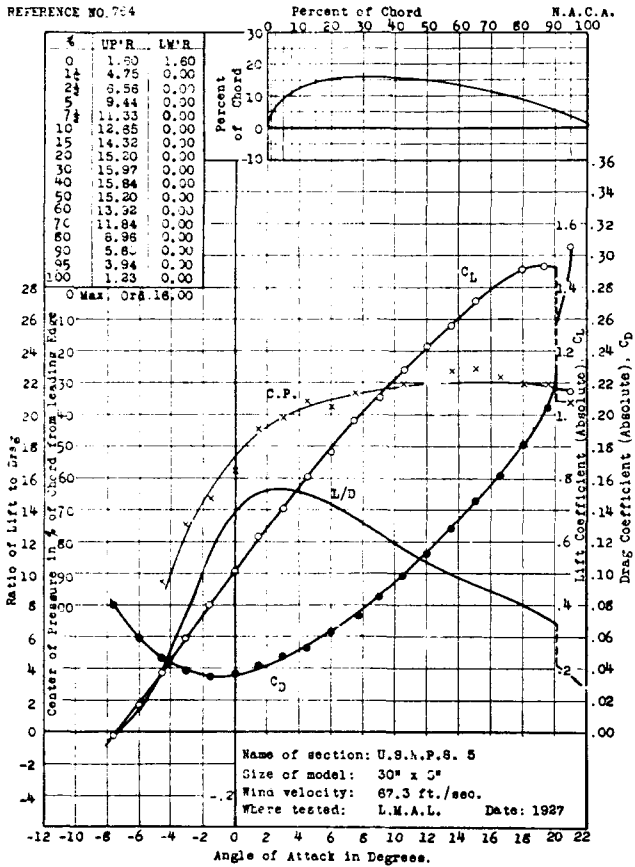
REFERENCE NO. 762



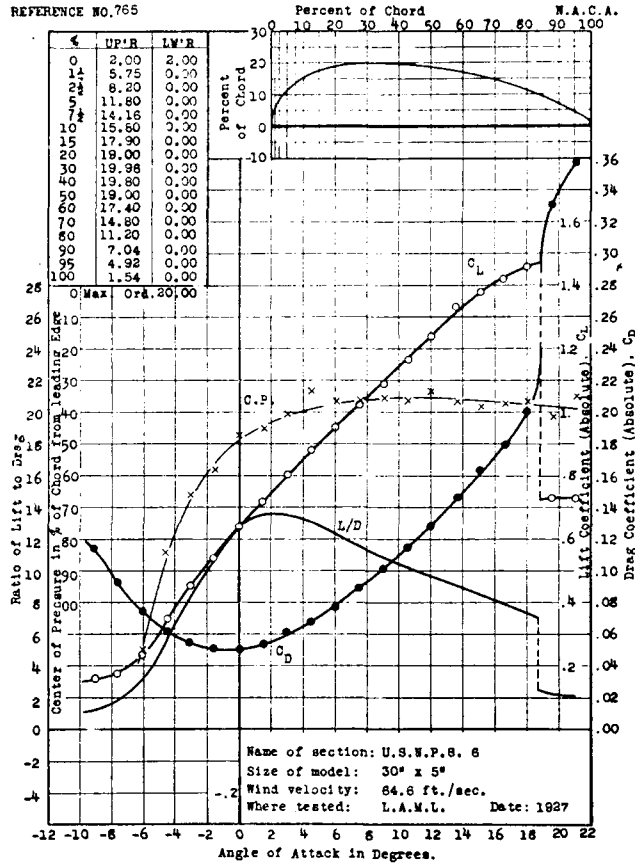
REFERENCE NO. 763



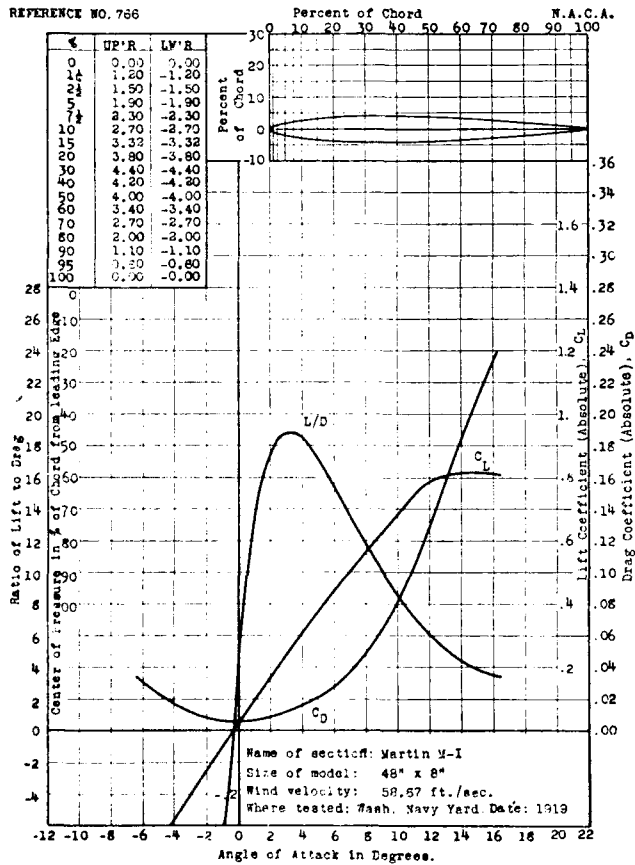
REFERENCE NO. 764



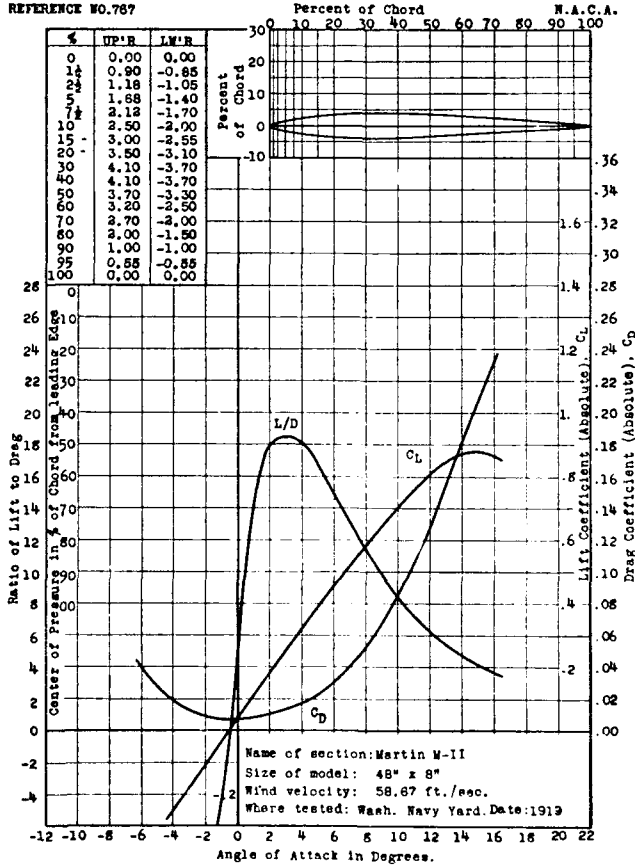
REFERENCE NO. 765

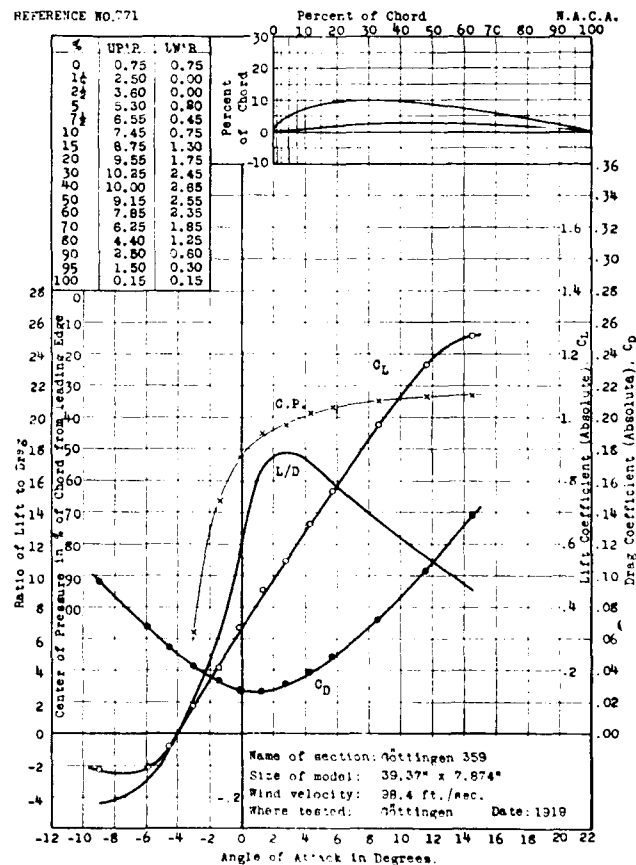
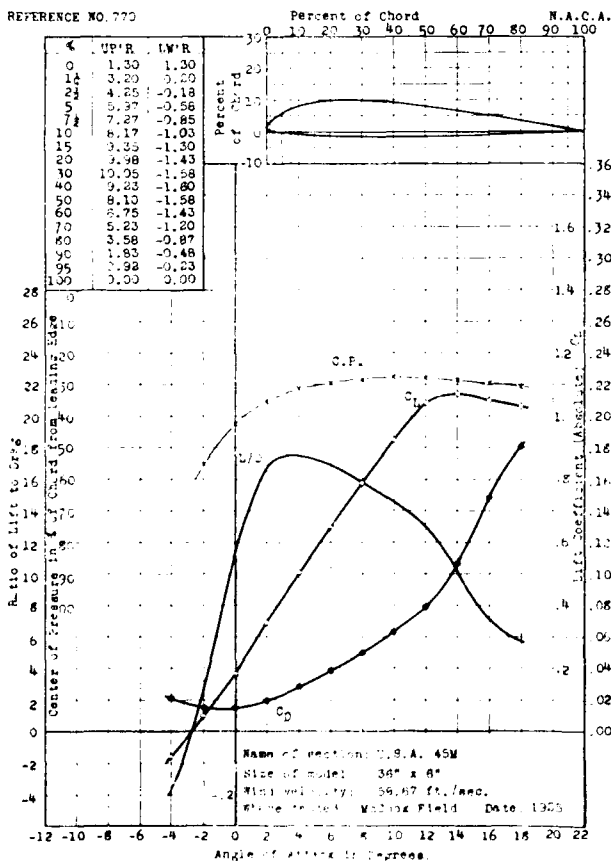
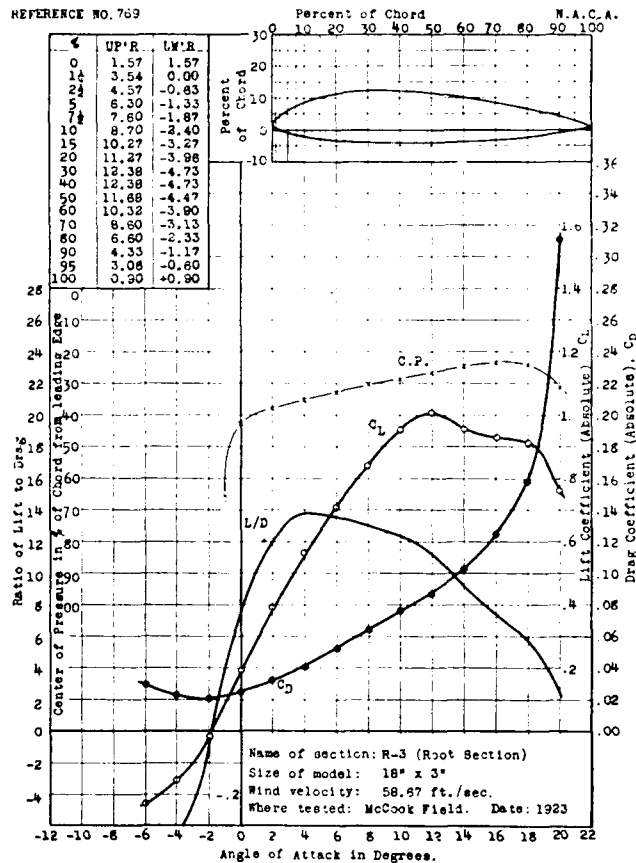
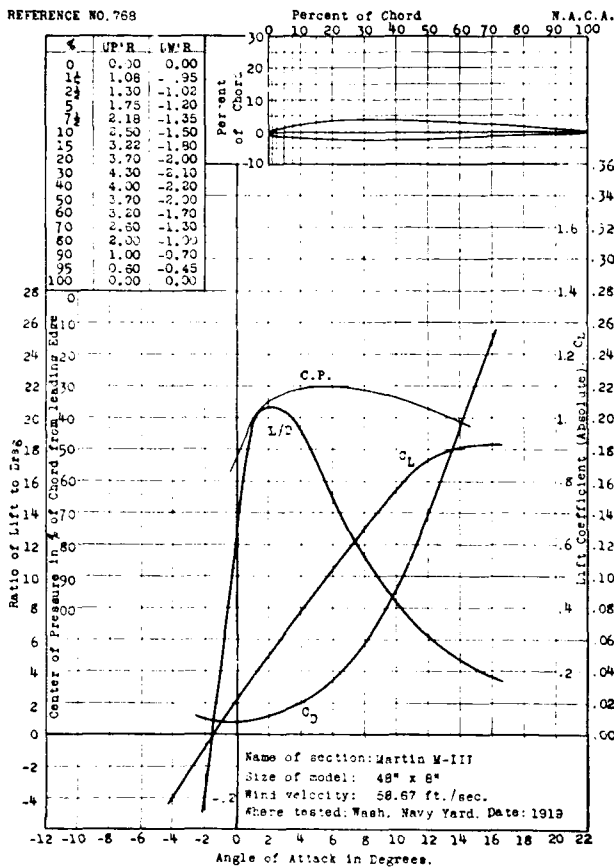


REFERENCE NO. 766

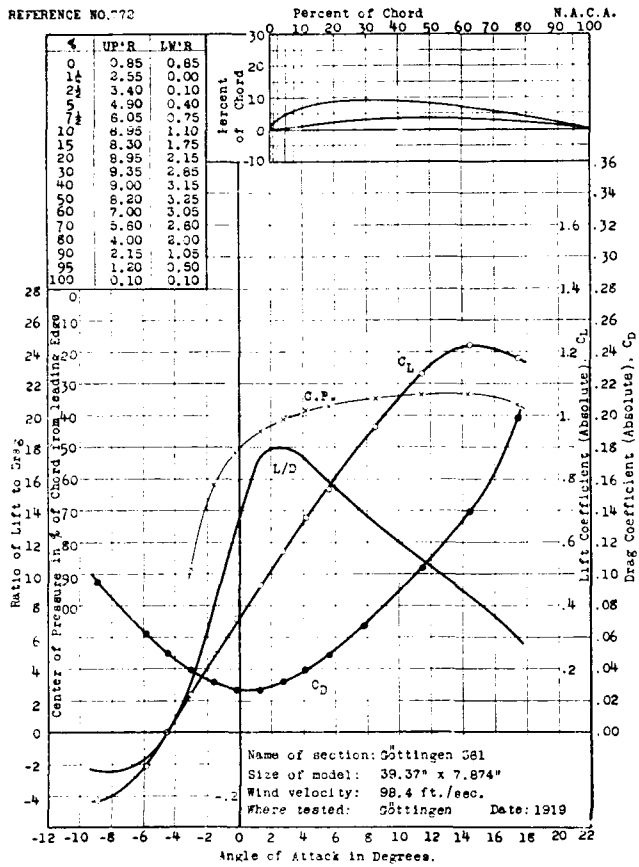


REFERENCE NO. 767

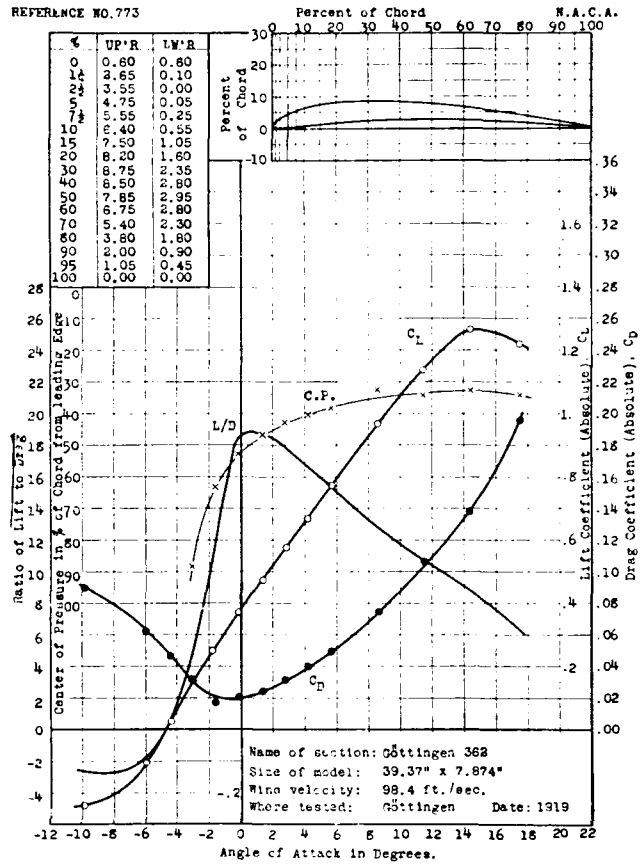




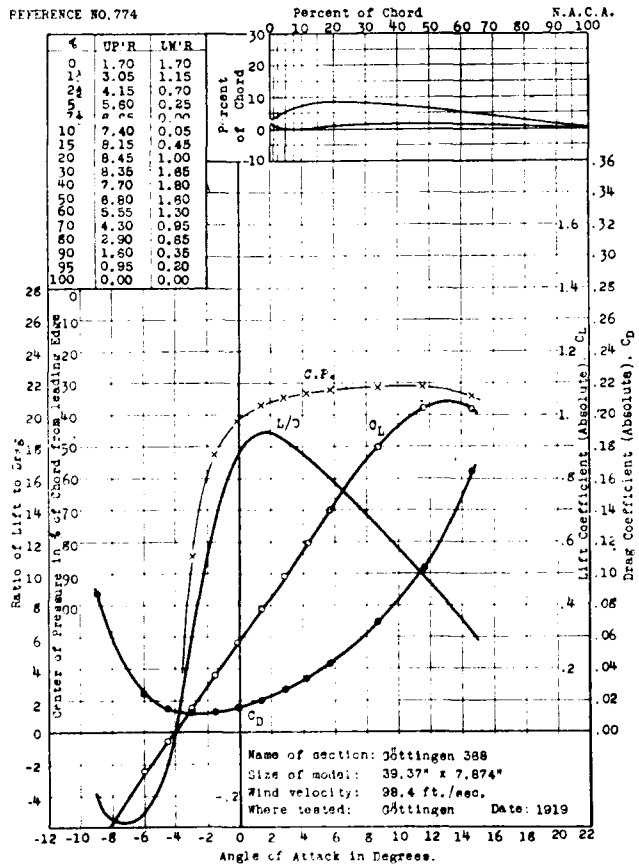
REFERENCE NO. 772



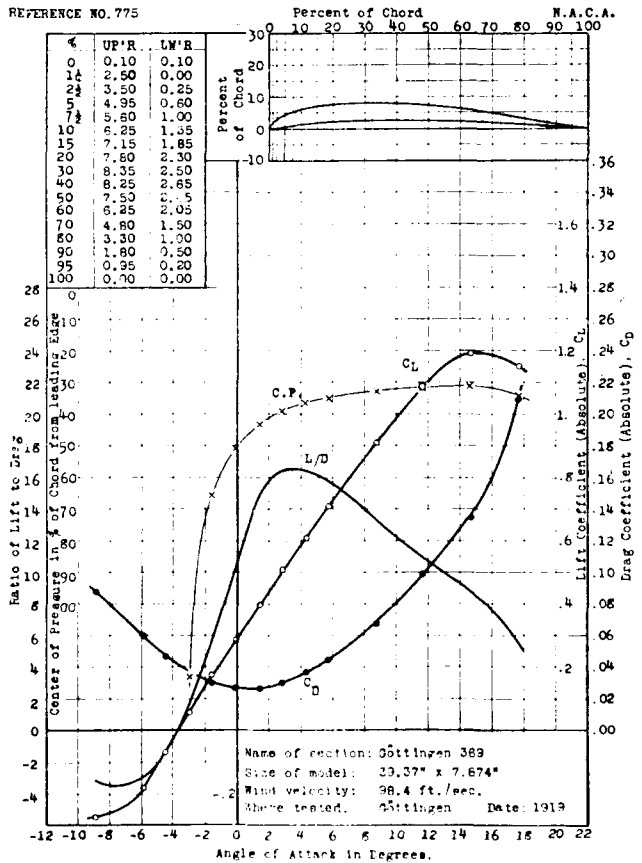
REFERENCE NO. 773



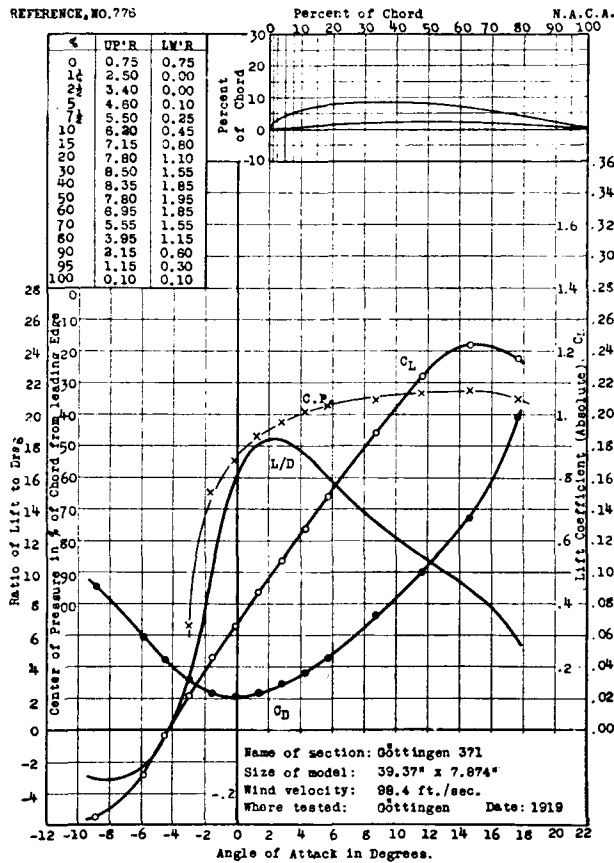
REFERENCE NO. 774



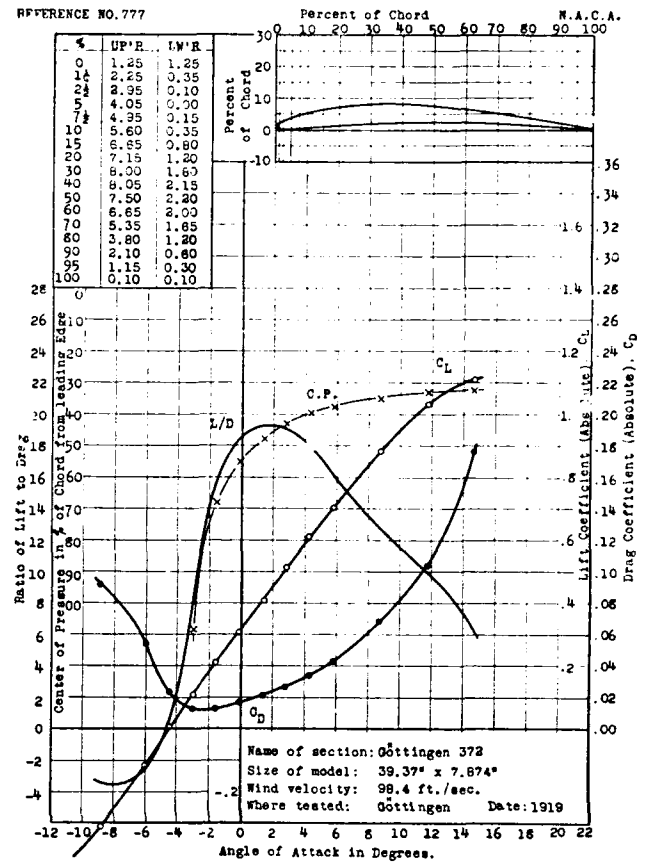
REFERENCE NO. 775



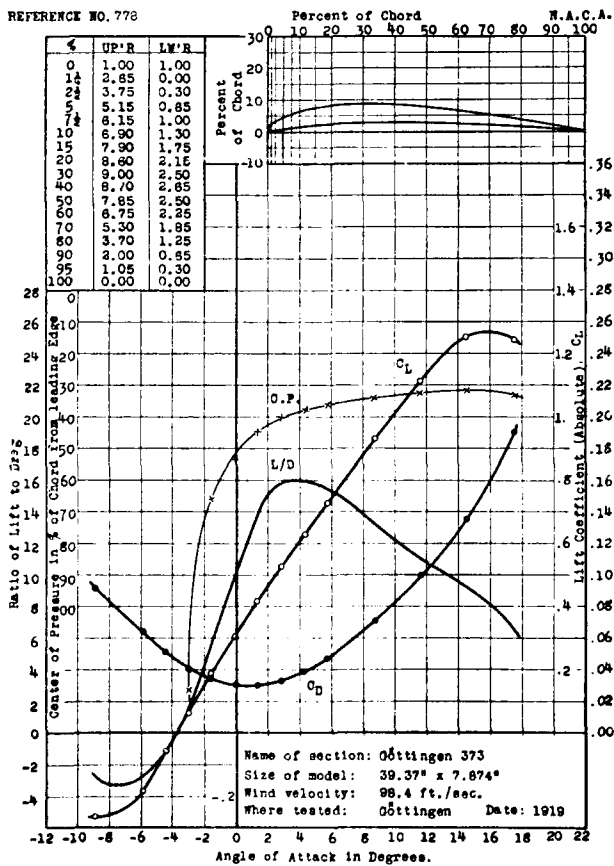
REFERENCE NO. 776



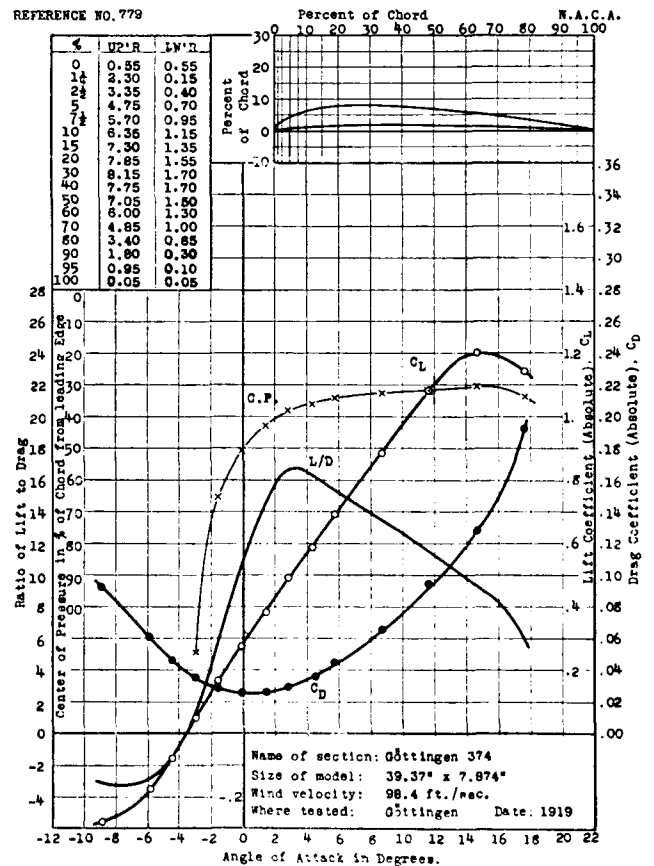
REFERENCE NO. 777

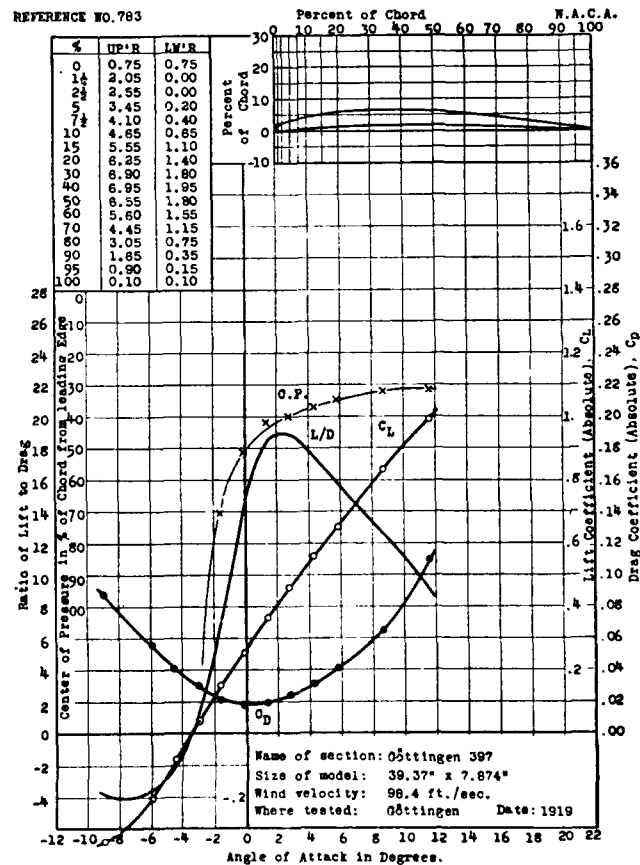
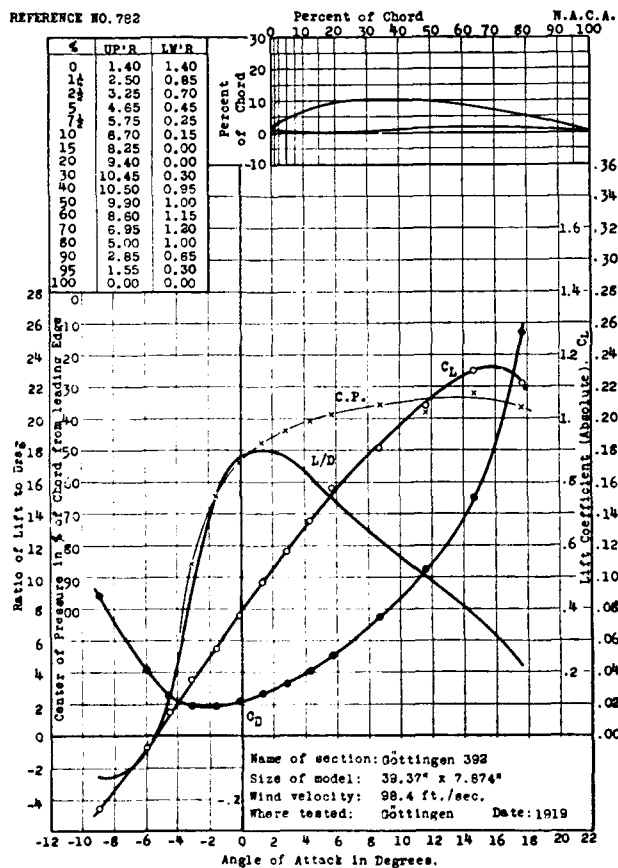
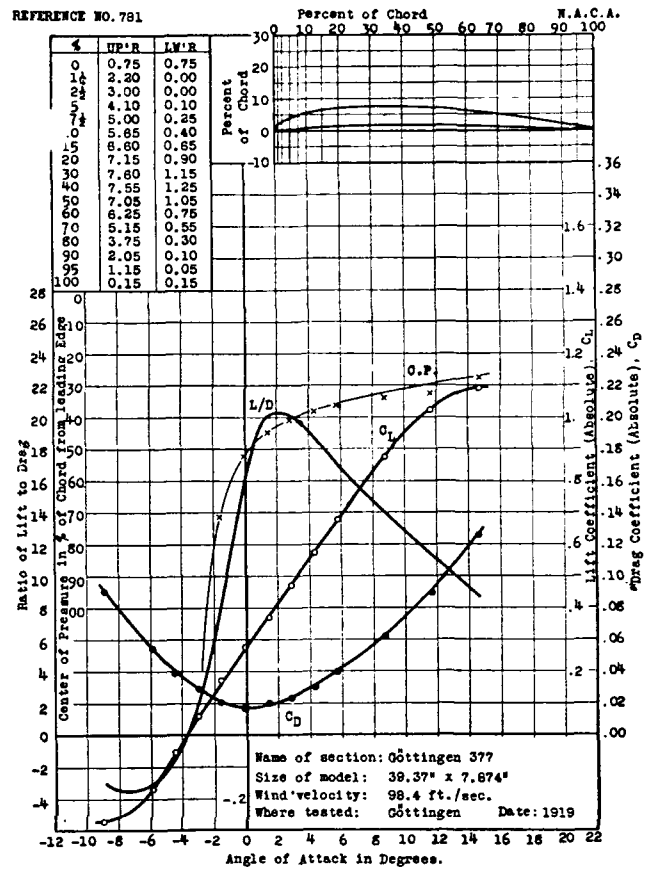
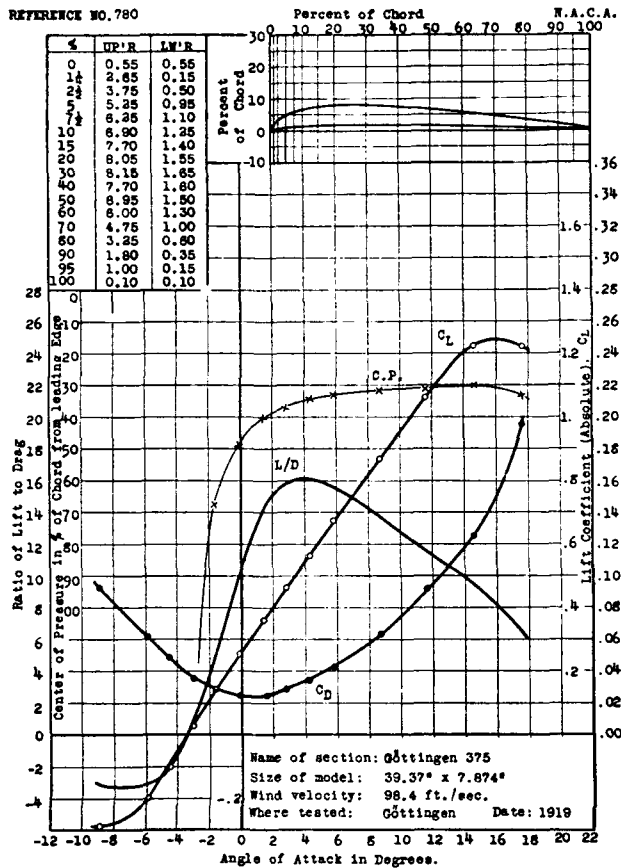


REFERENCE NO. 778

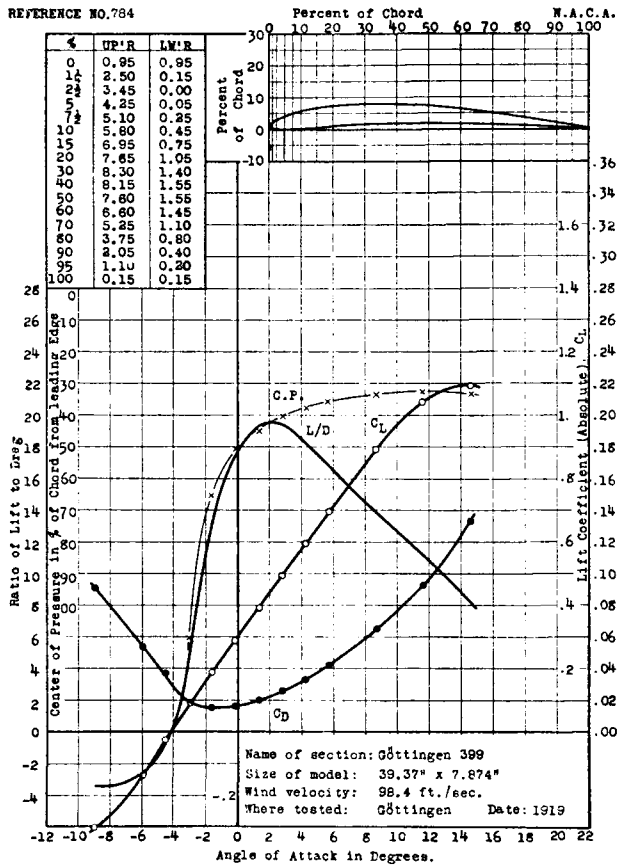


REFERENCE NO. 779

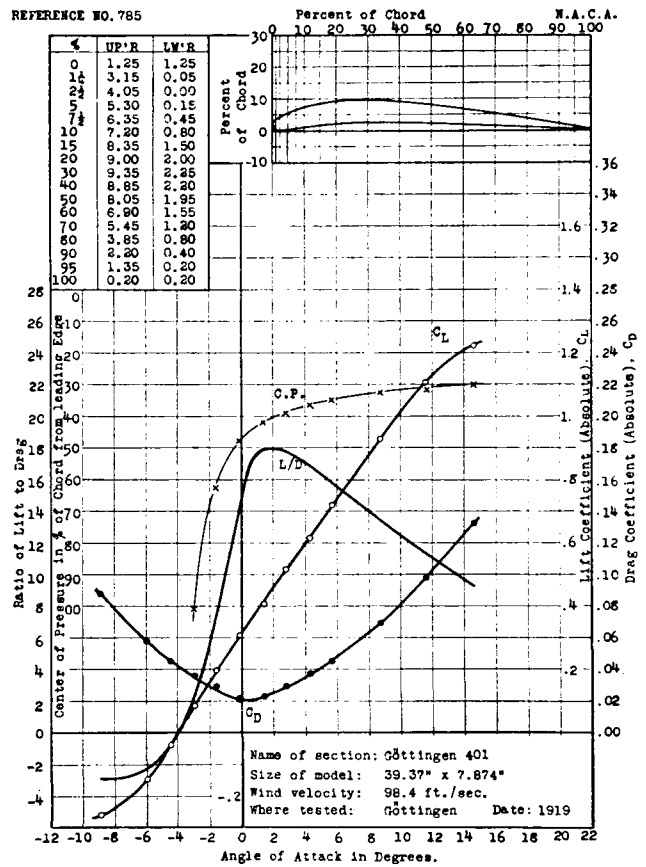




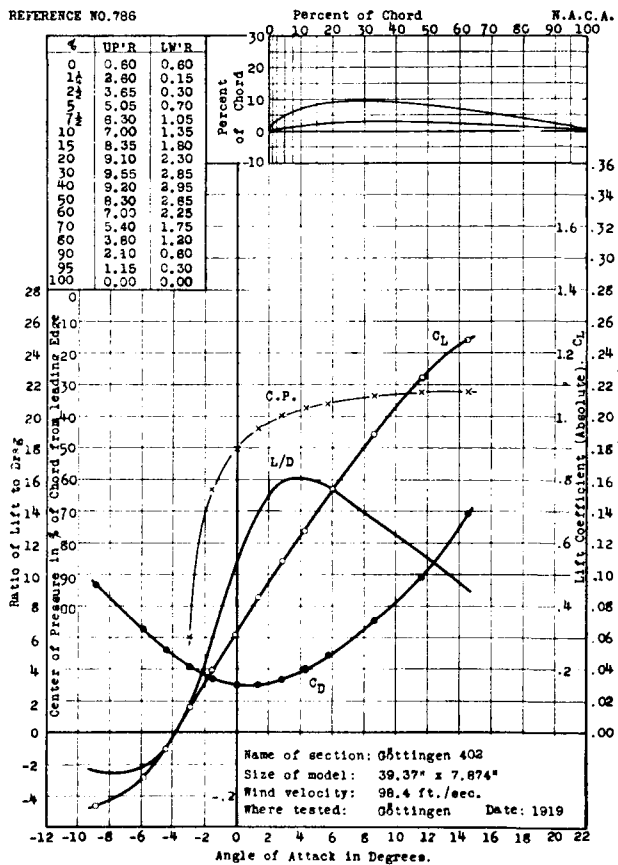
REFERENCE NO. 784



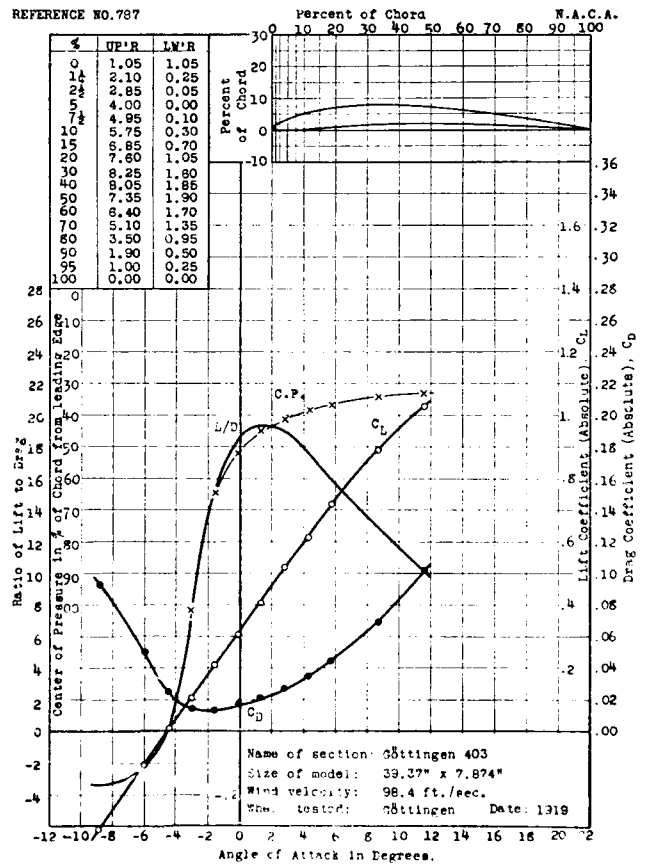
REFERENCE NO. 785

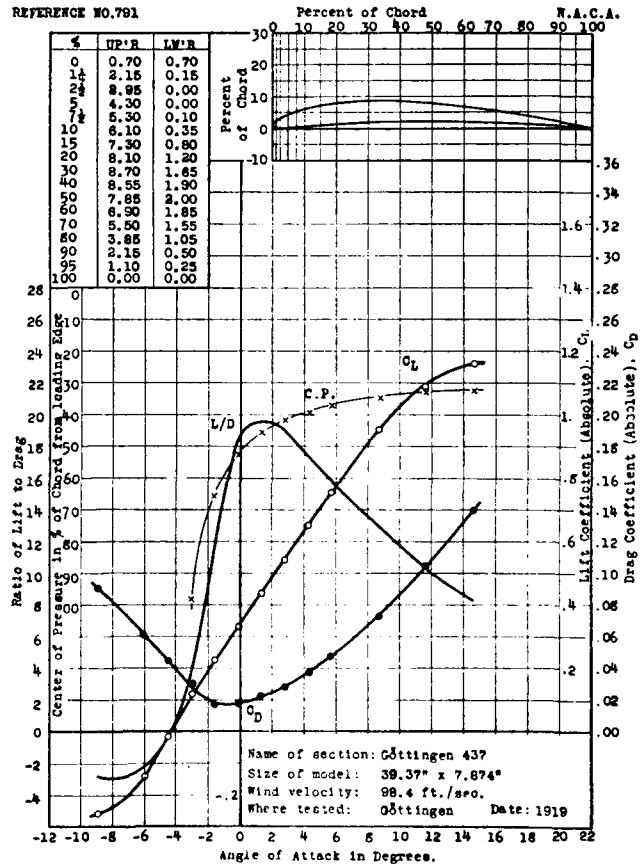
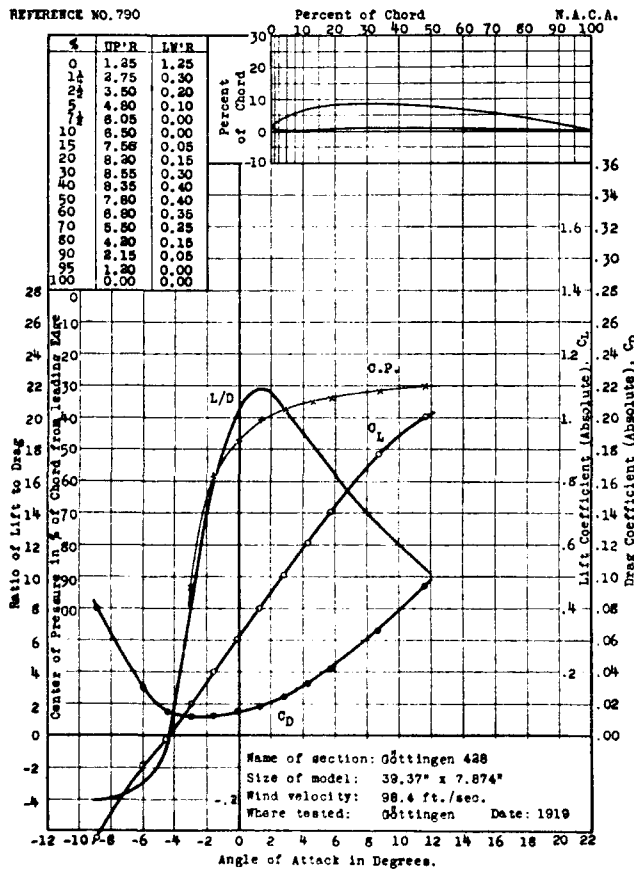
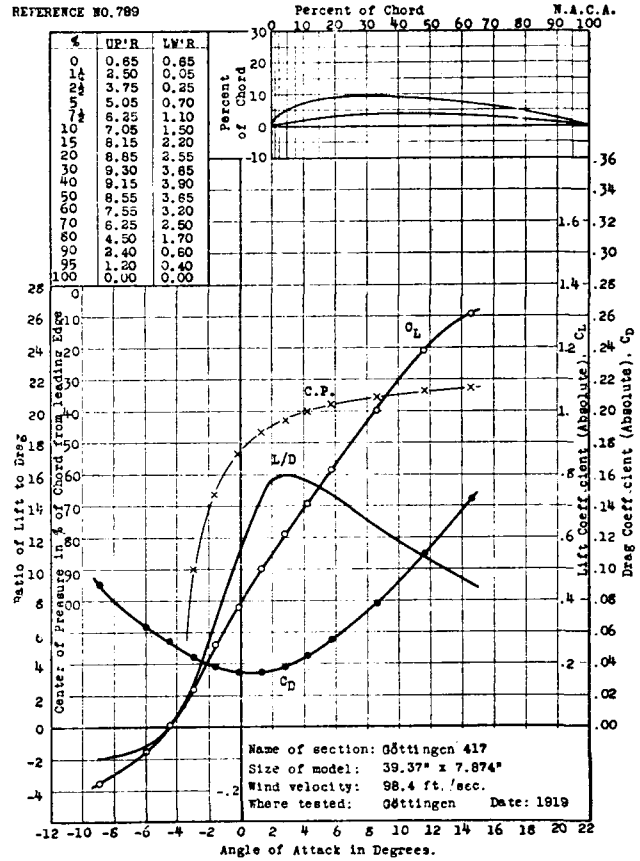
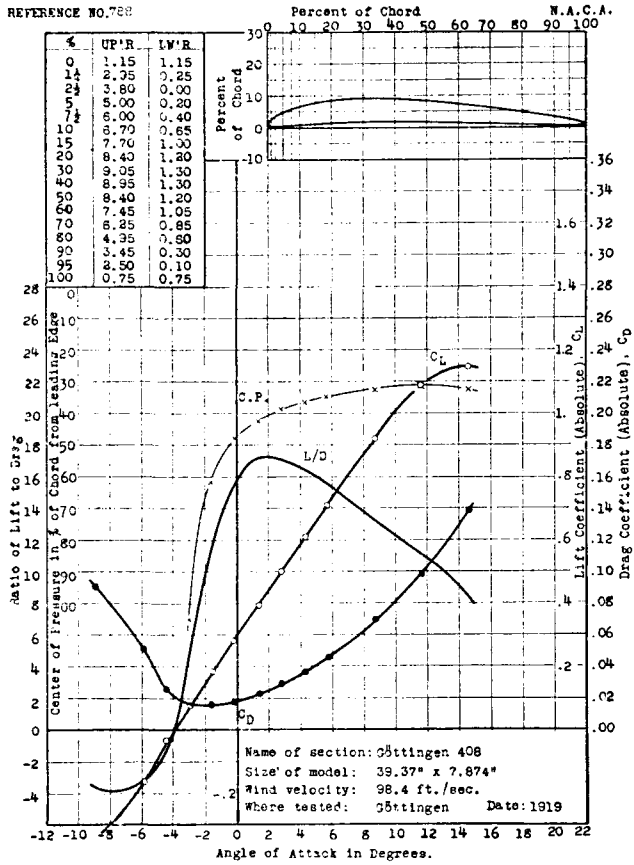


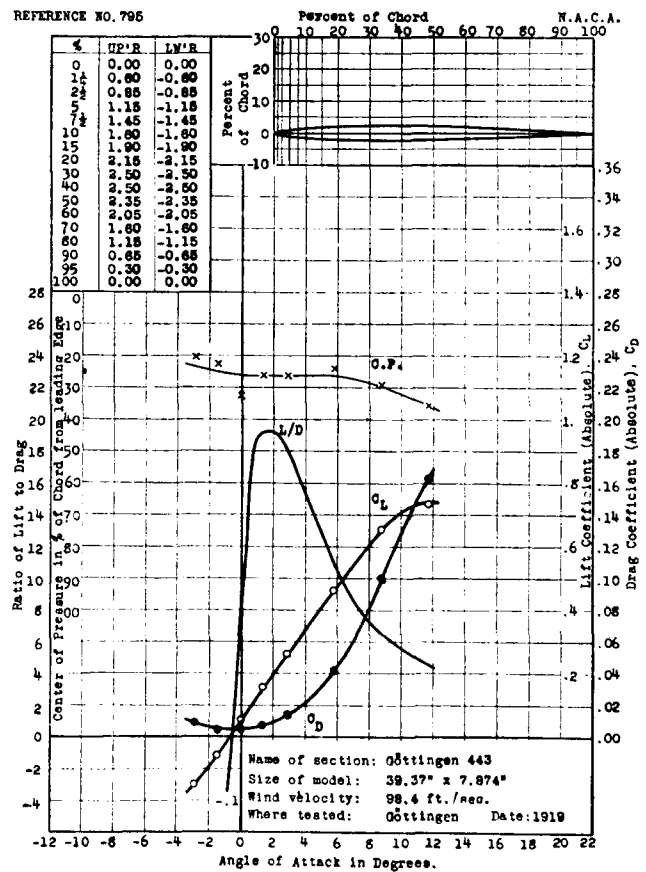
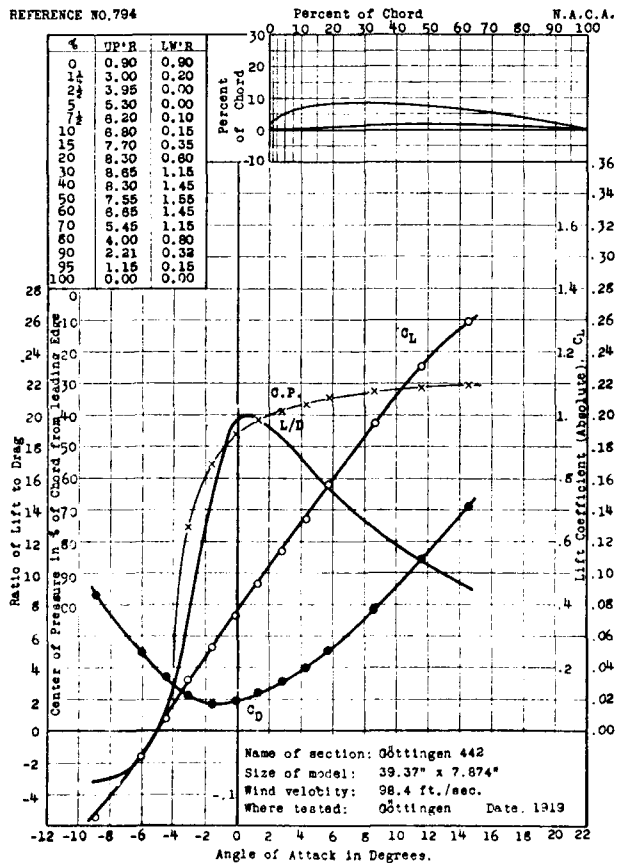
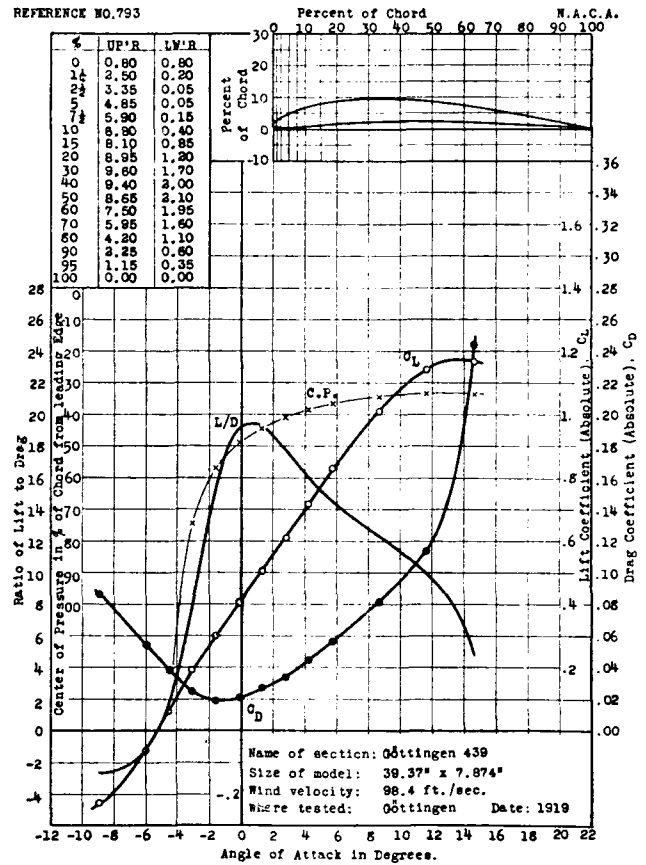
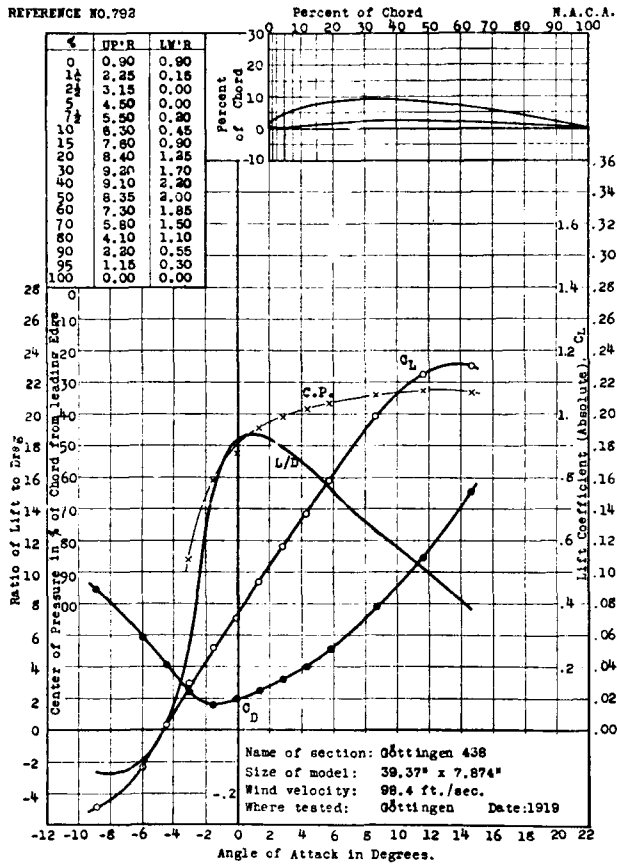
REFERENCE NO. 786



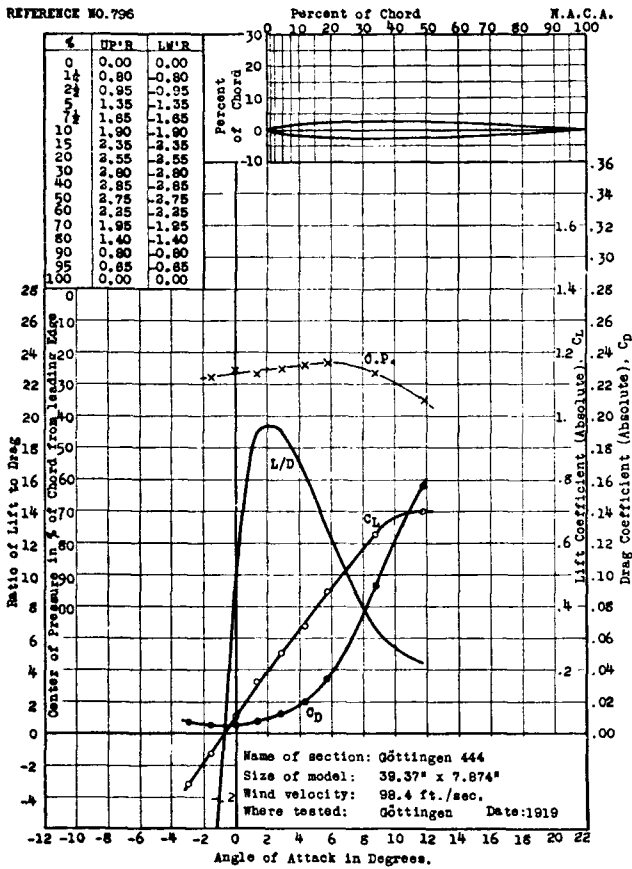
REFERENCE NO. 787



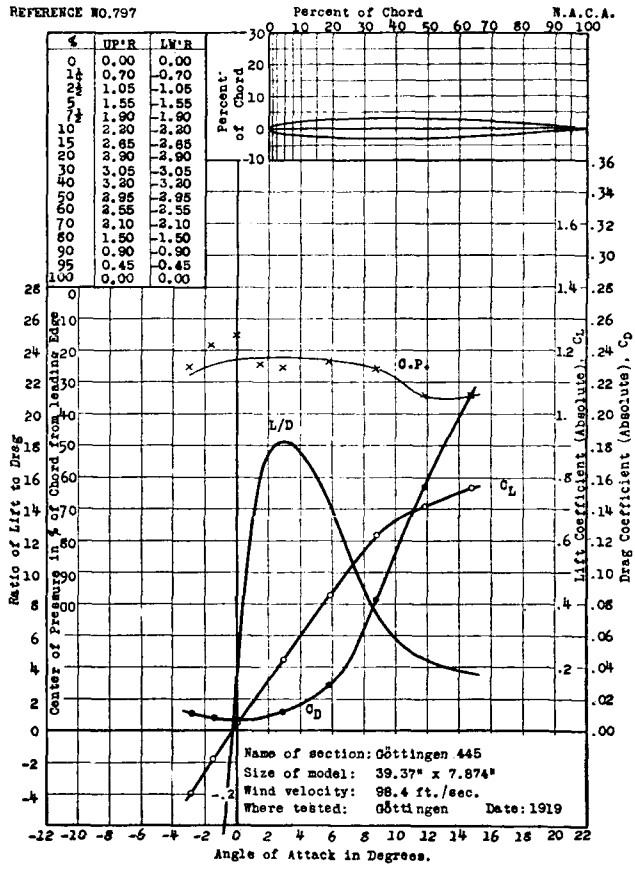




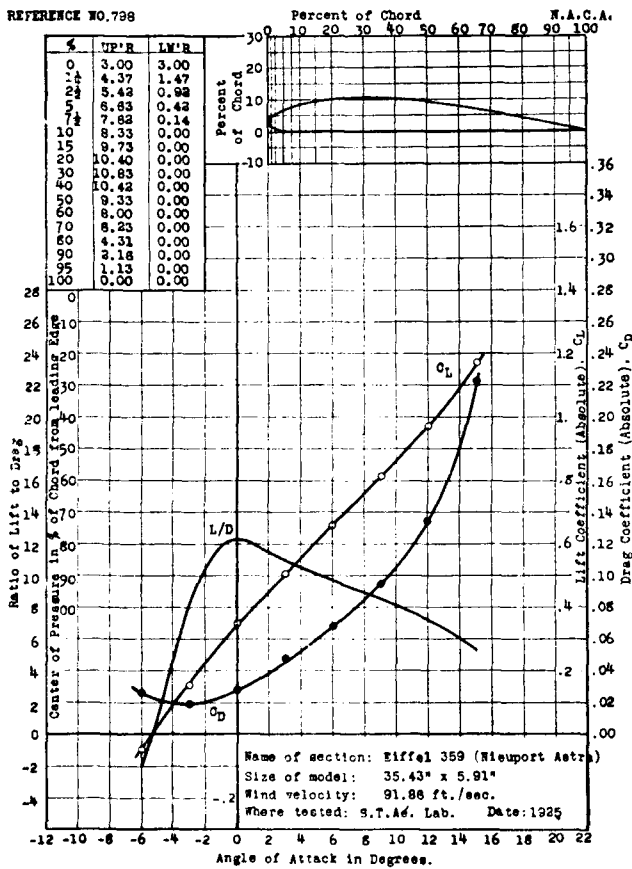
REFERENCE NO. 796



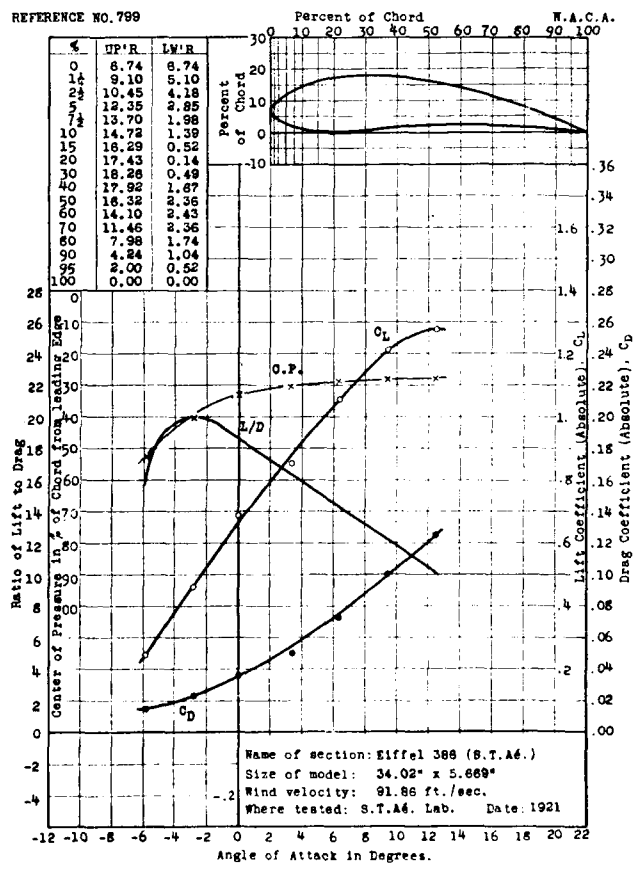
REFERENCE NO. 797



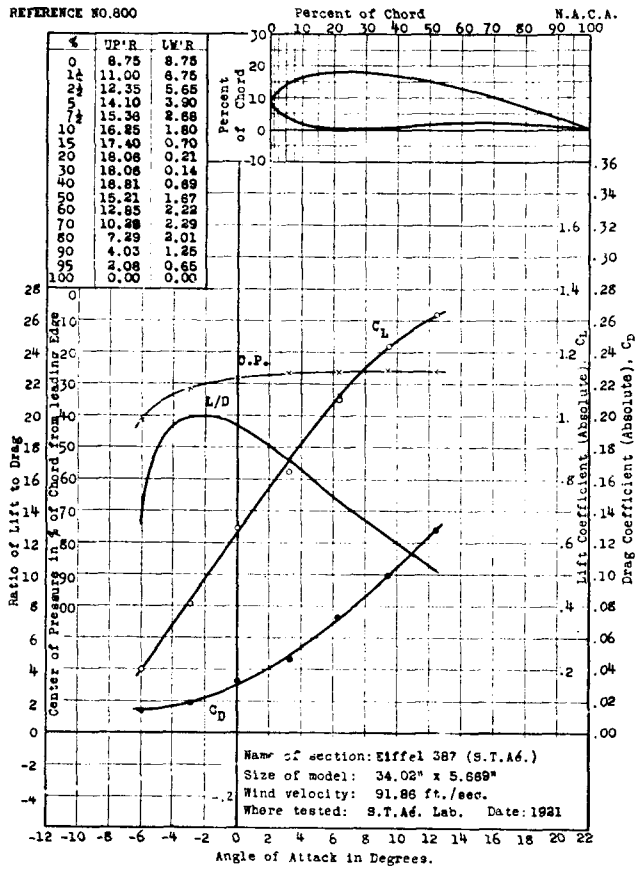
REFERENCE NO. 798



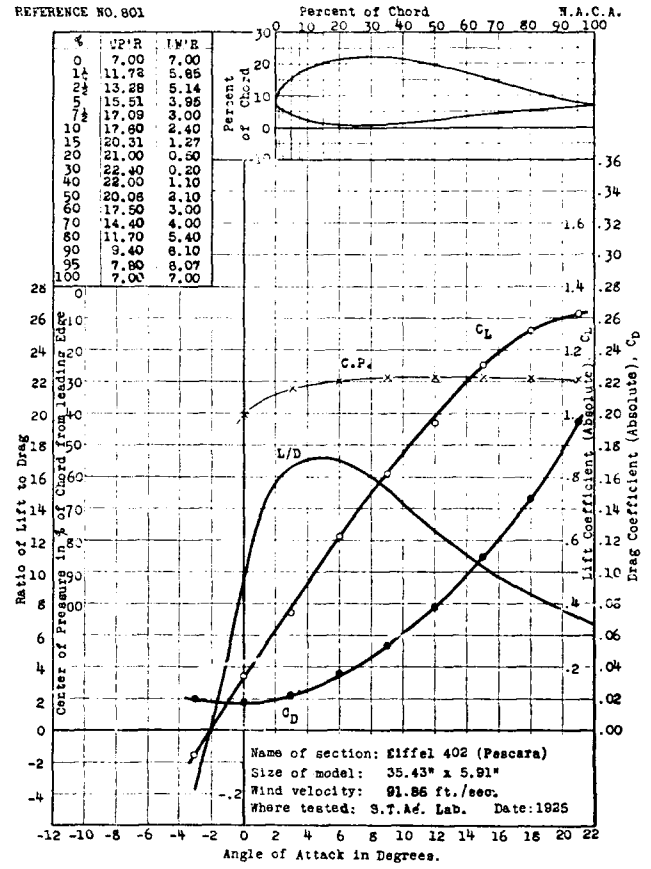
REFERENCE NO. 799



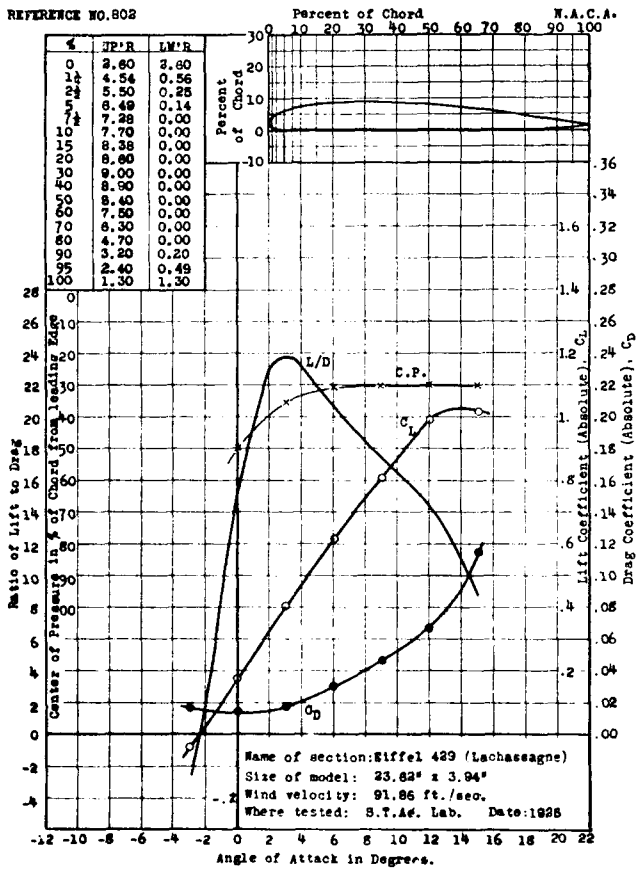
REFERENCE NO. 800



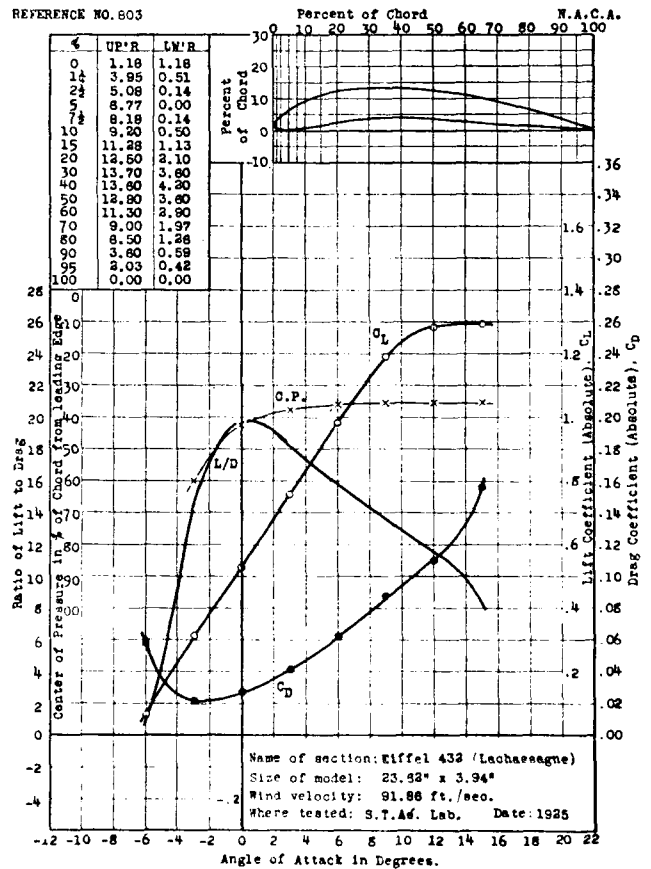
REFERENCE NO. 801

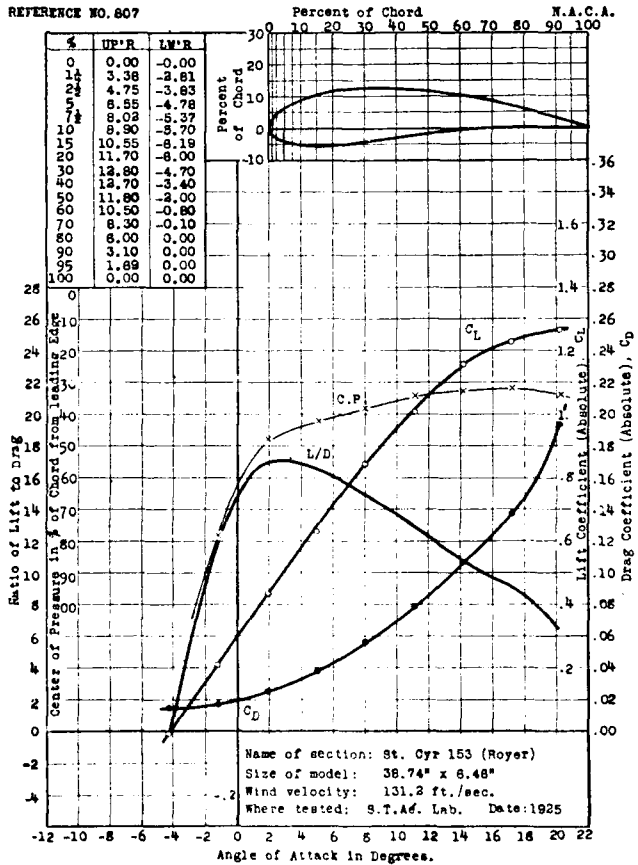
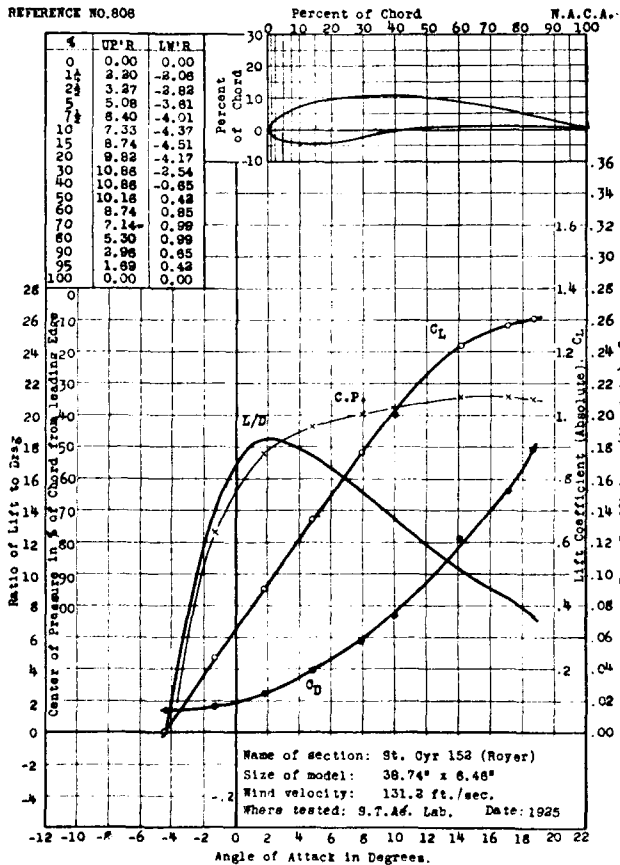
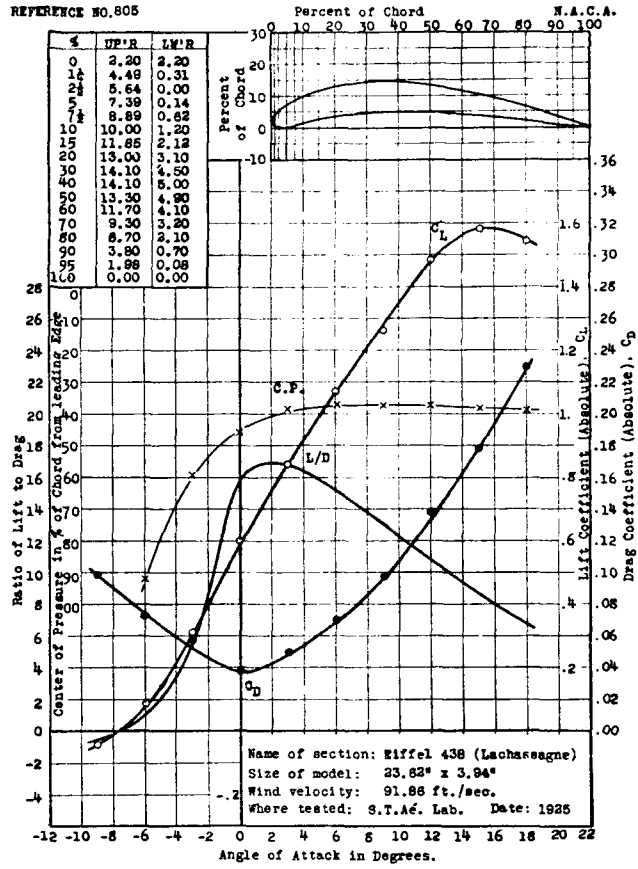
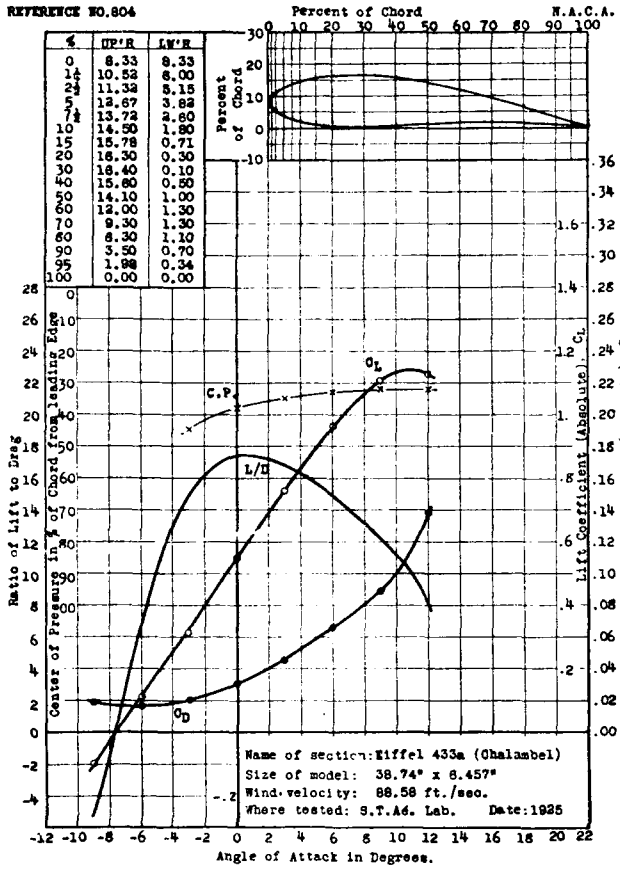


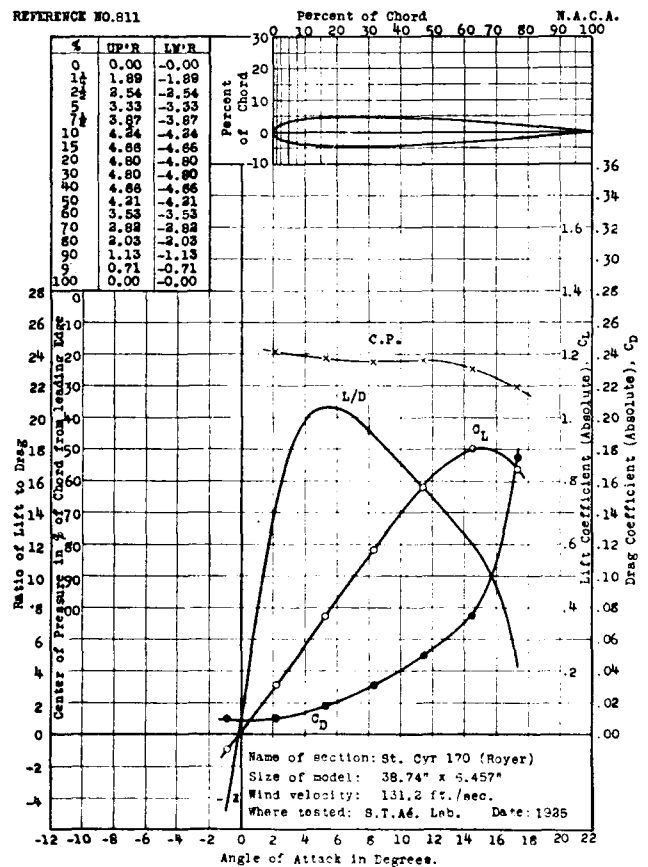
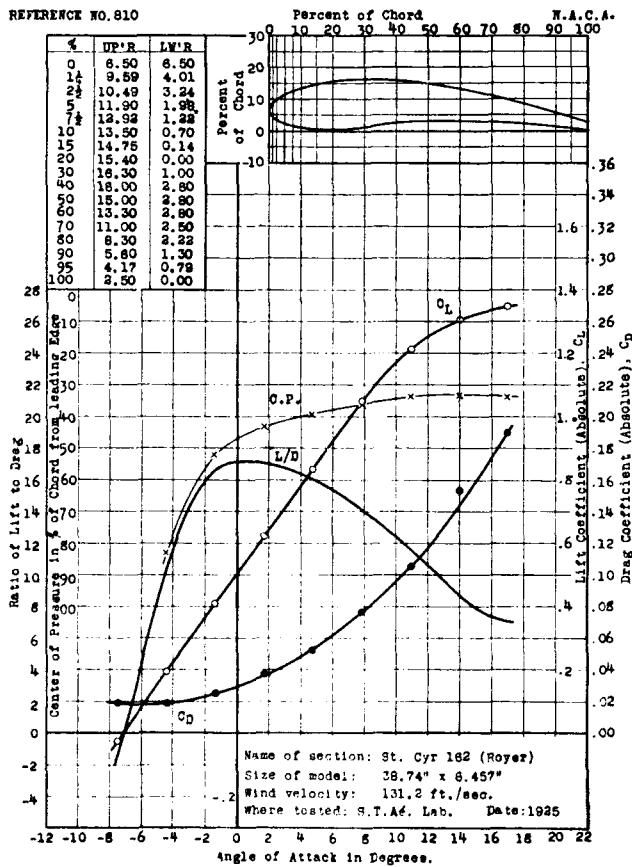
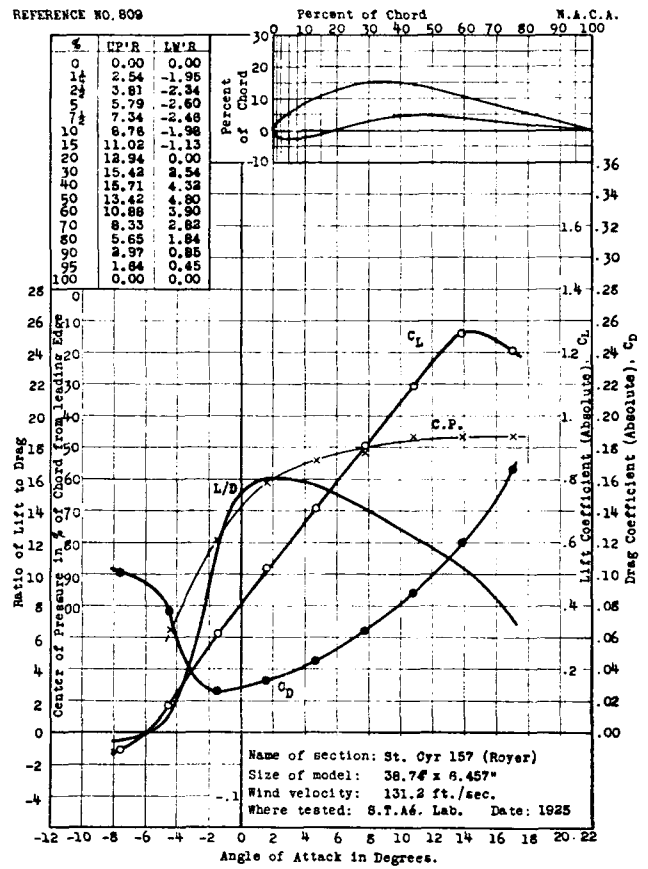
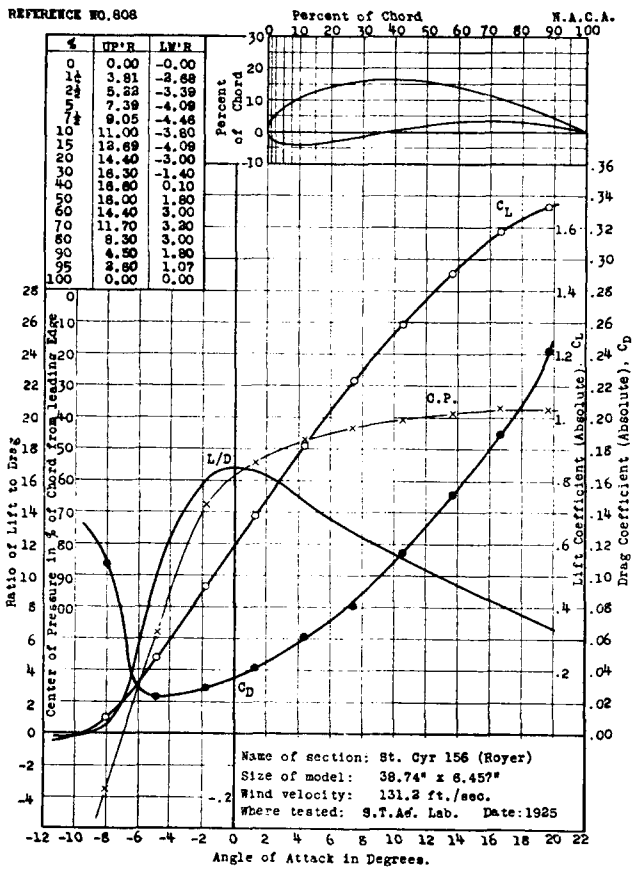
REFERENCE NO. 802



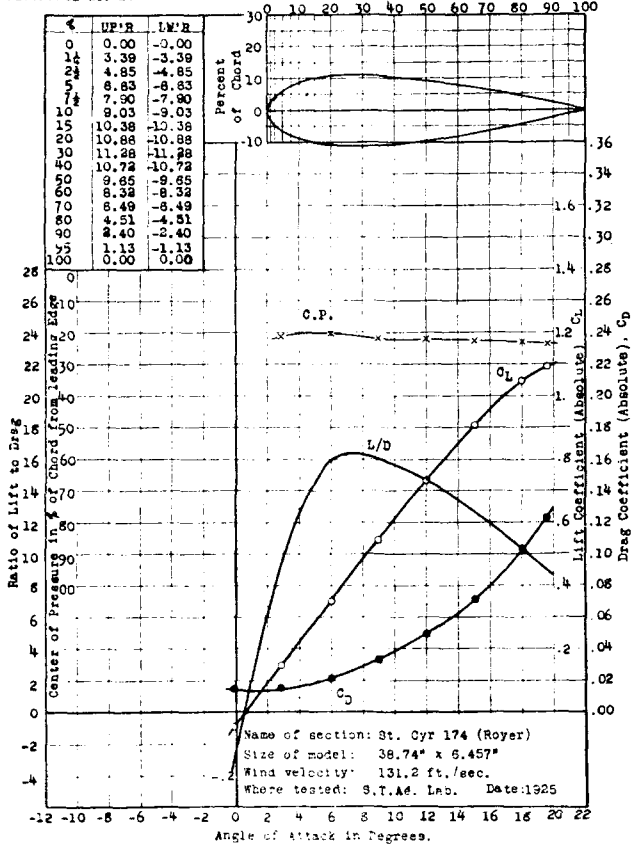
REFERENCE NO. 803



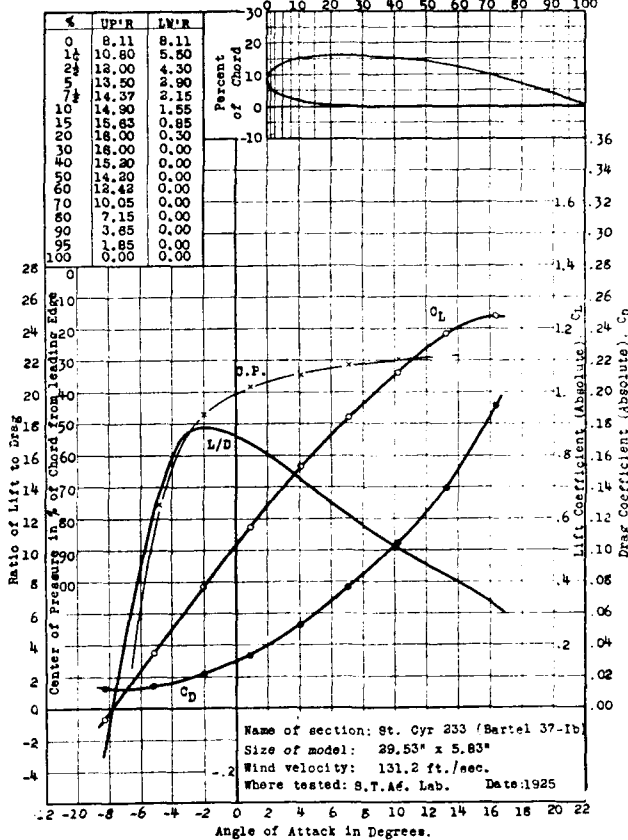




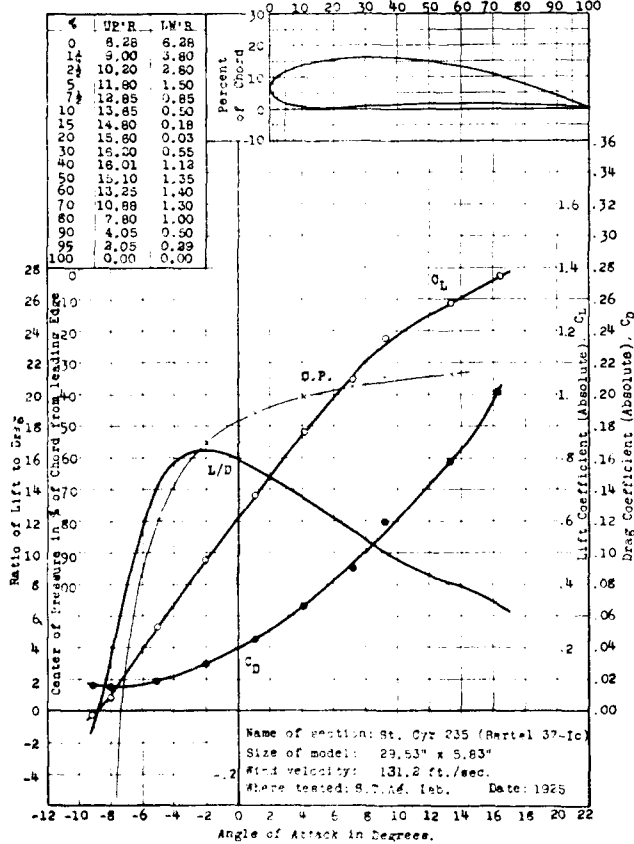
REFERENCE NO. 813



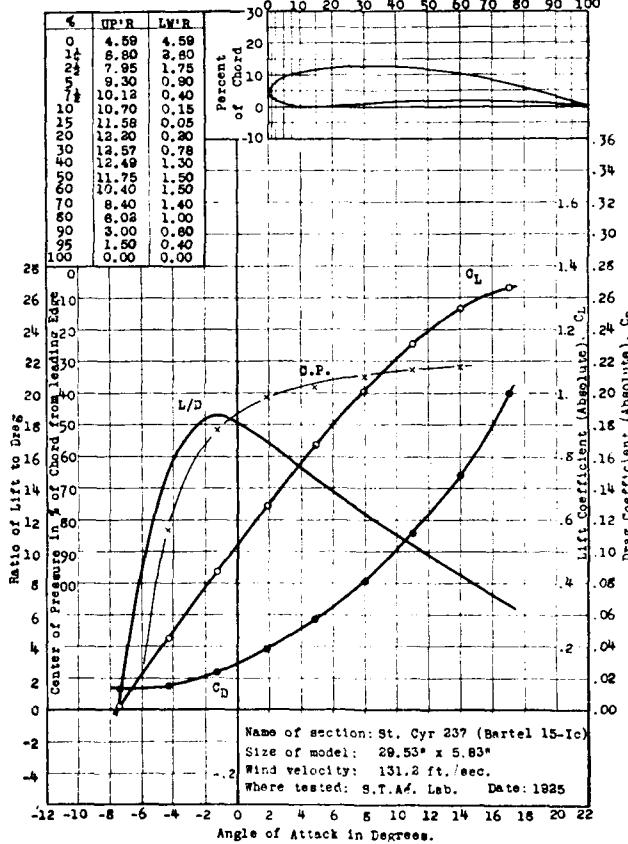
REFERENCE NO. 813

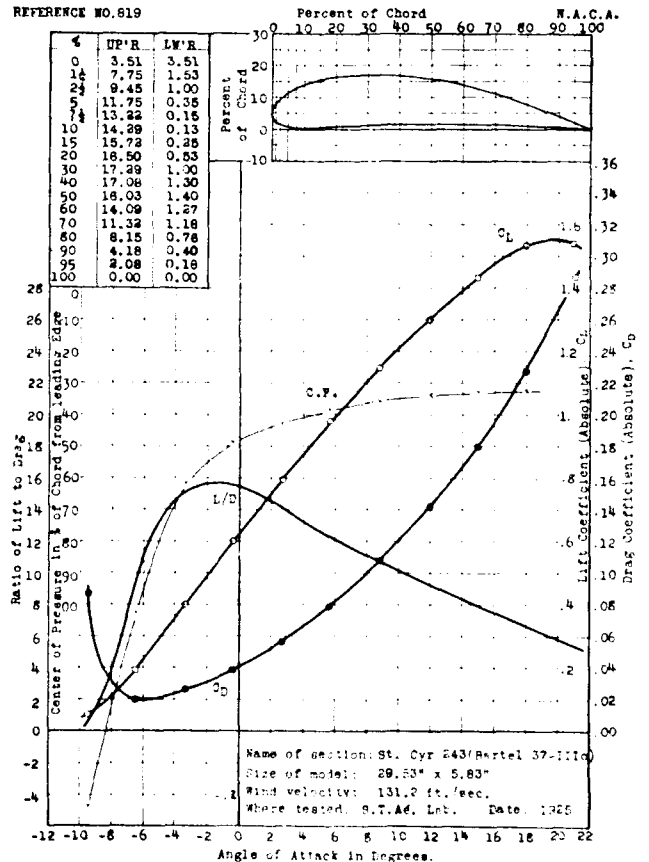
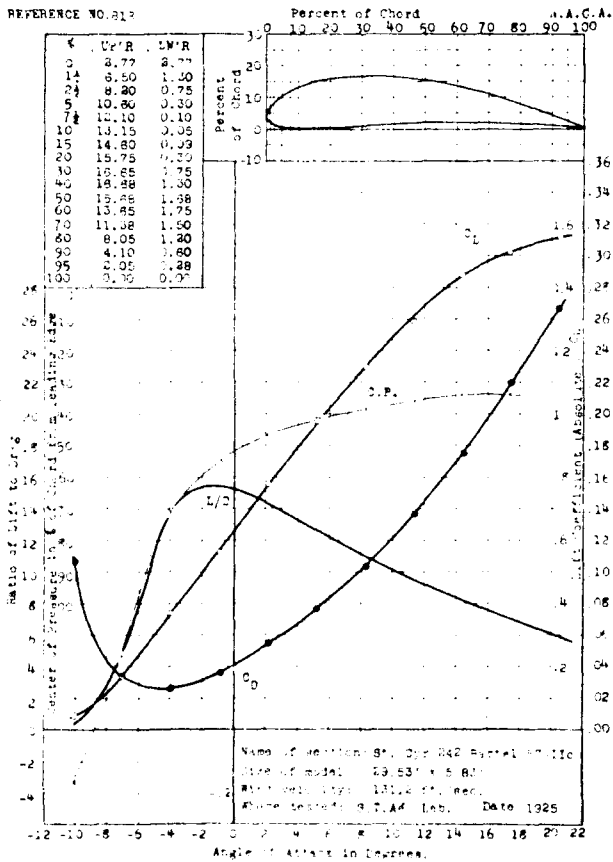
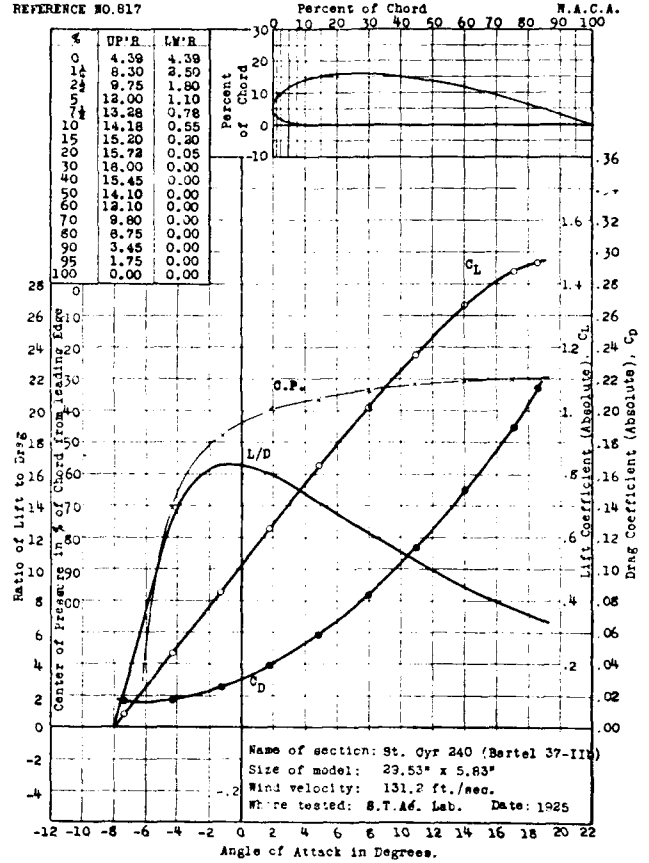
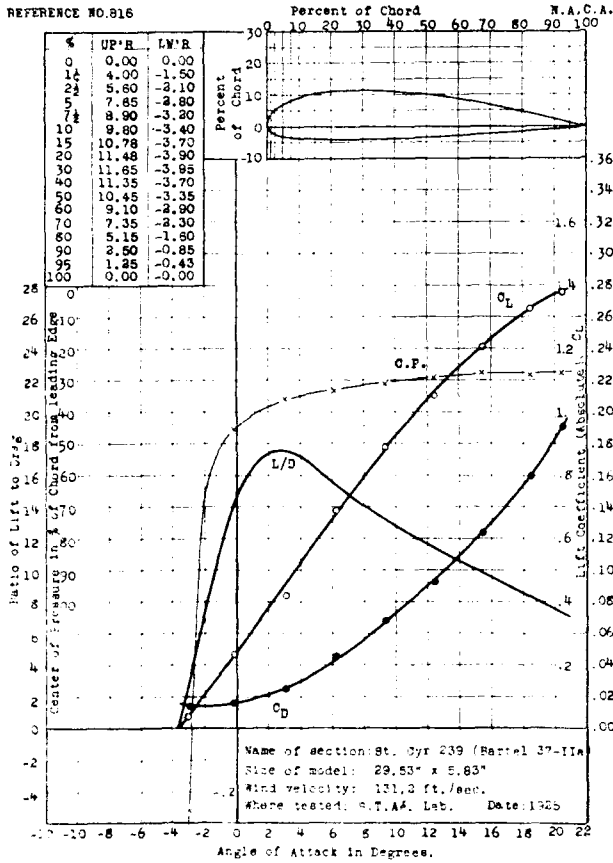


REFERENCE NO. 814

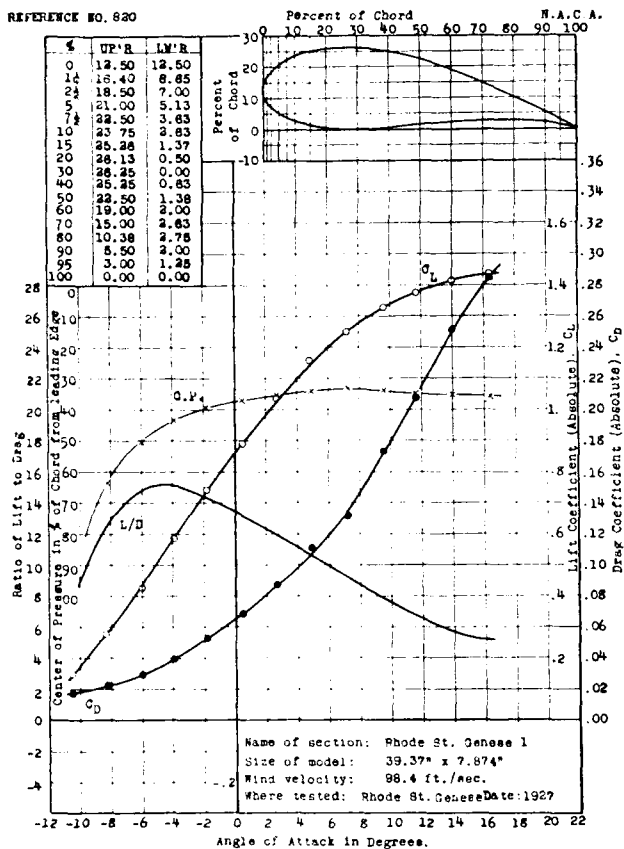


REFERENCE NO. 815

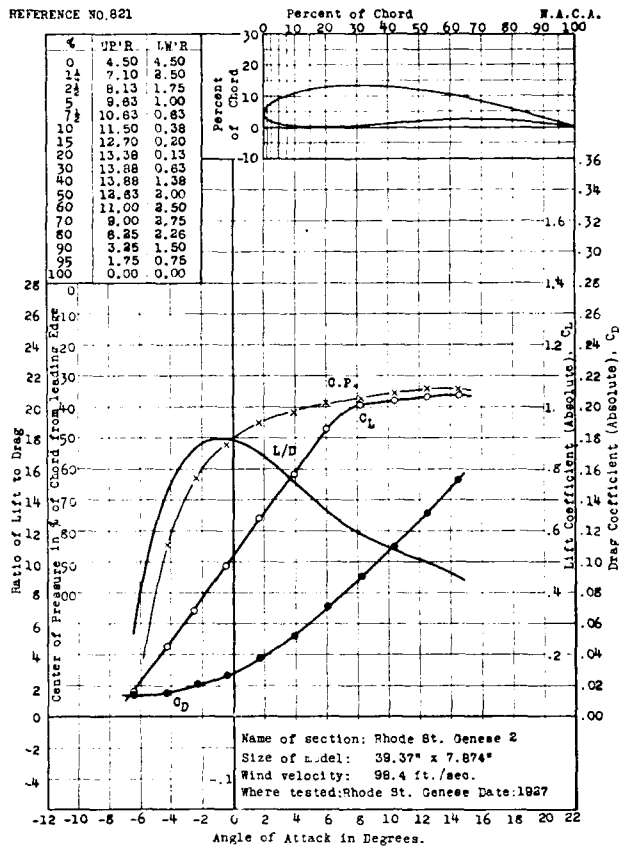




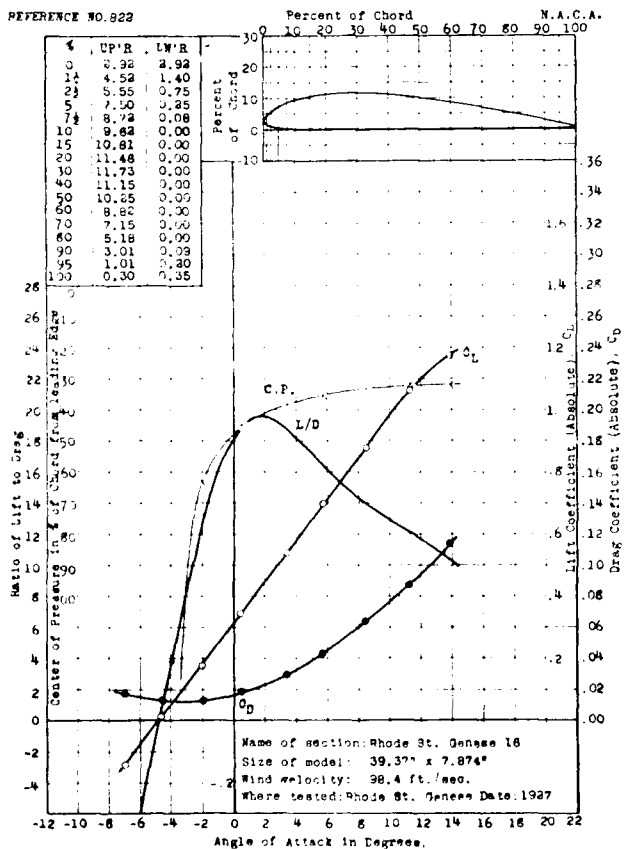
REFERENCE NO. 820



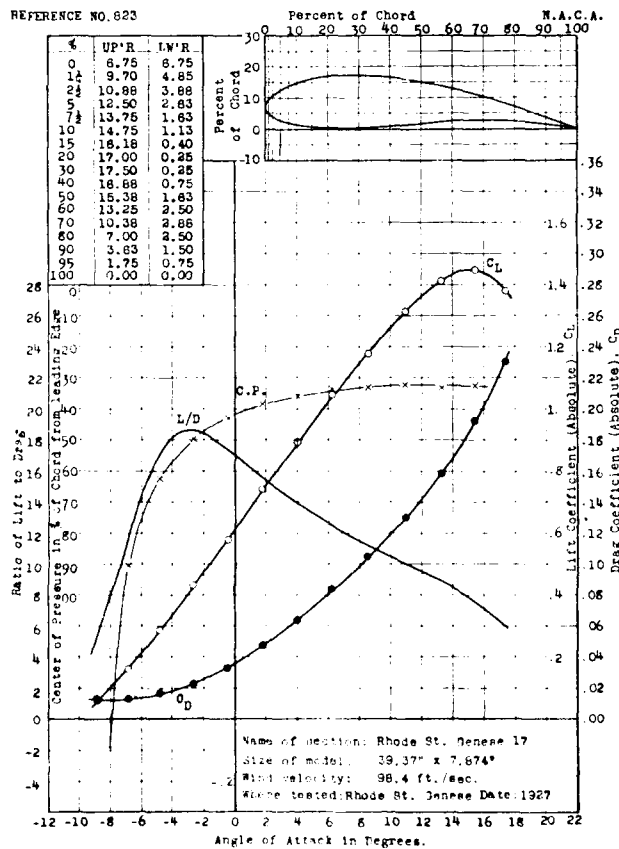
REFERENCE NO. 821



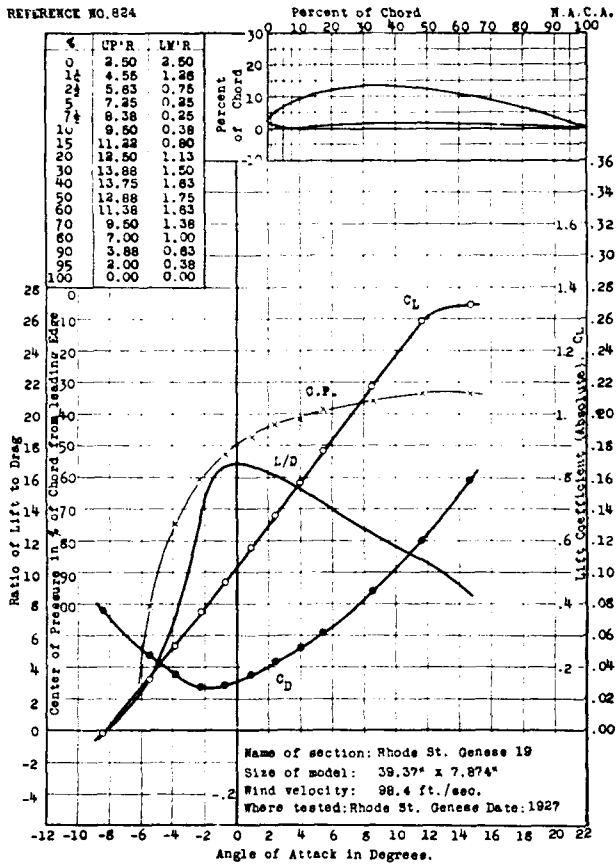
REFERENCE NO. 822



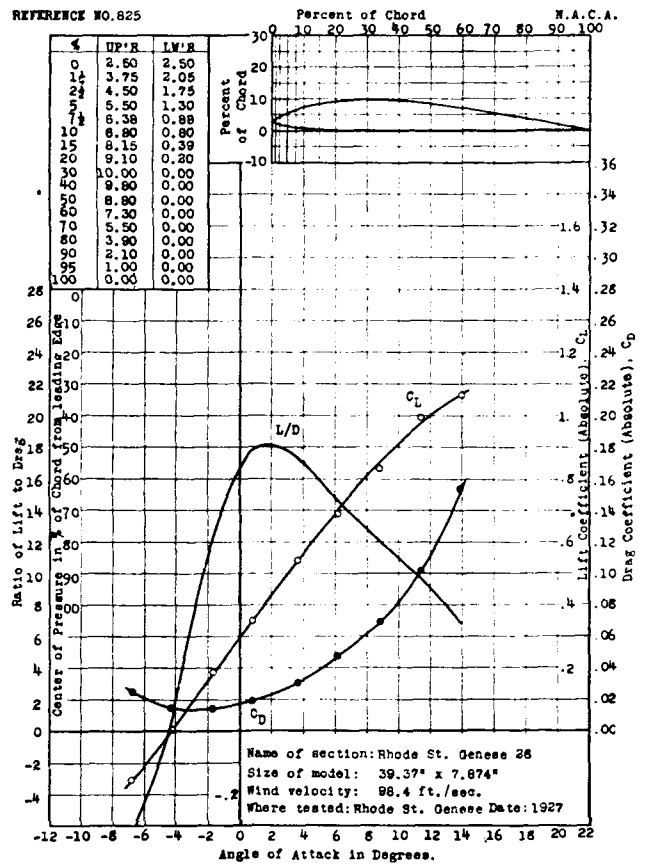
REFERENCE NO. 823



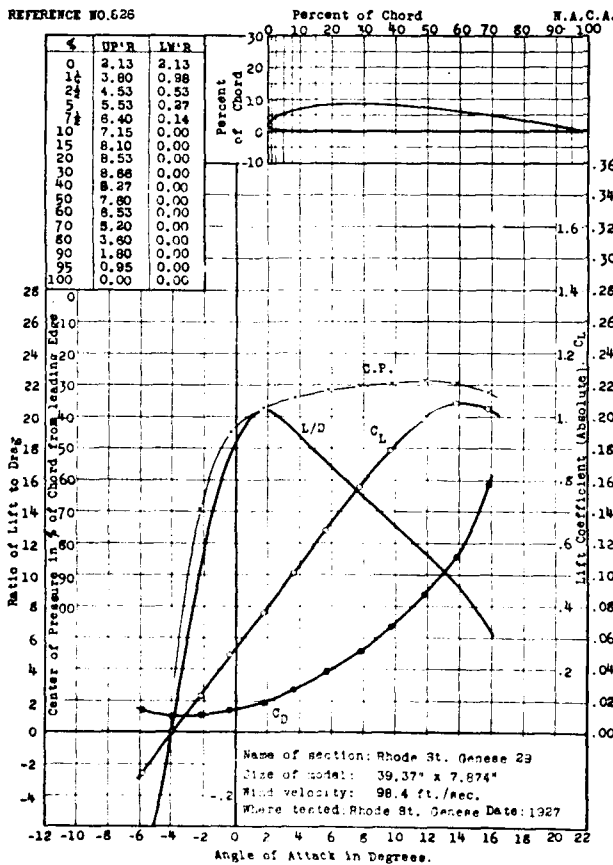
REFERENCE NO. 824



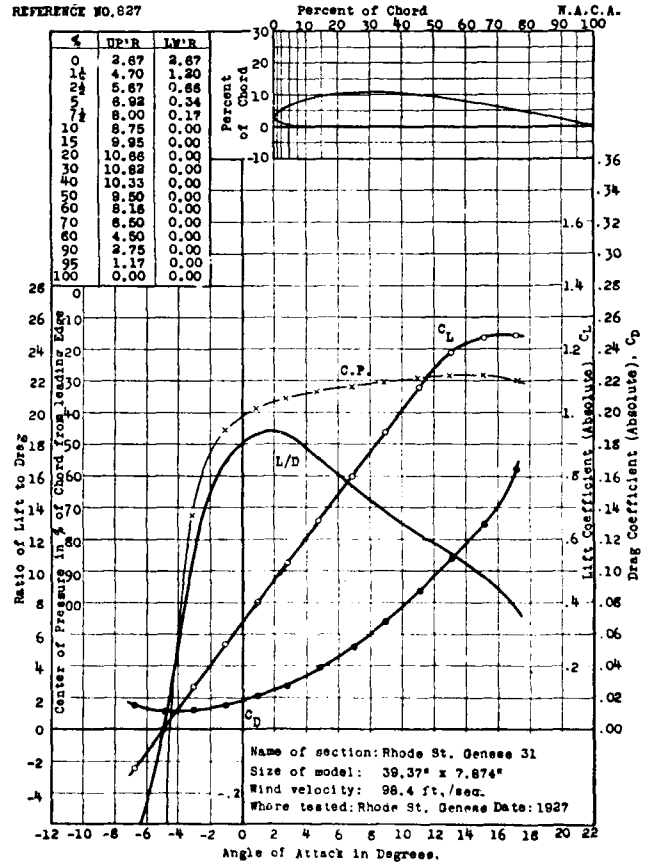
REFERENCE NO. 825



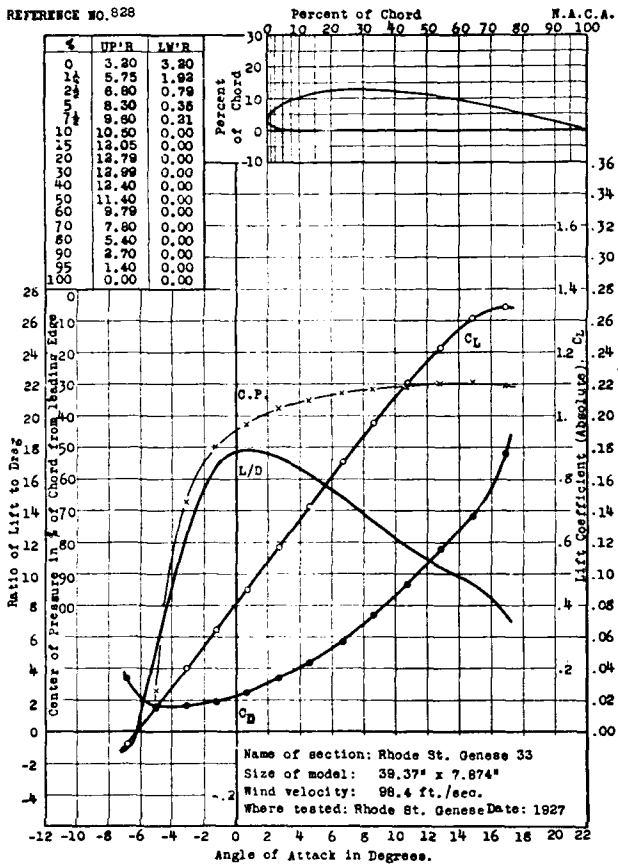
REFERENCE NO. 826



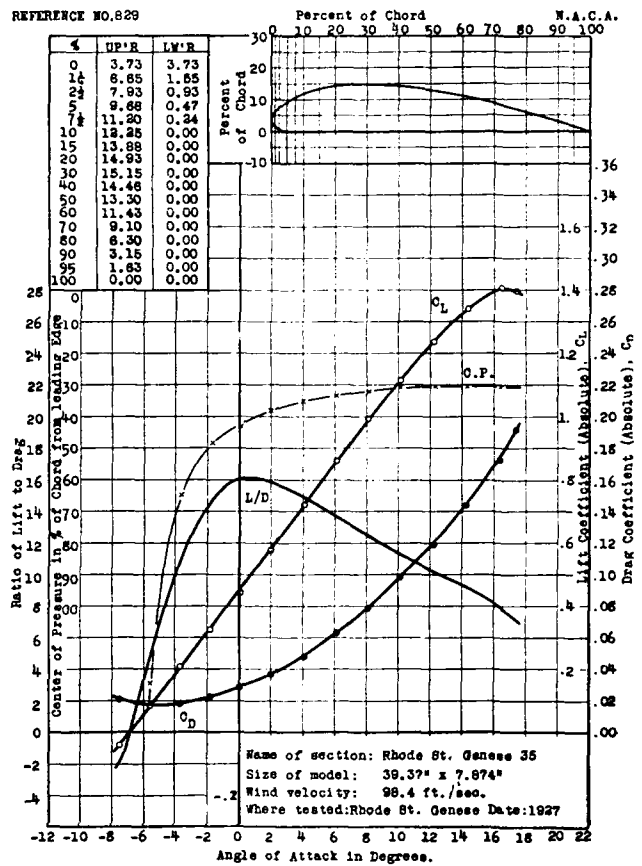
REFERENCE NO. 827



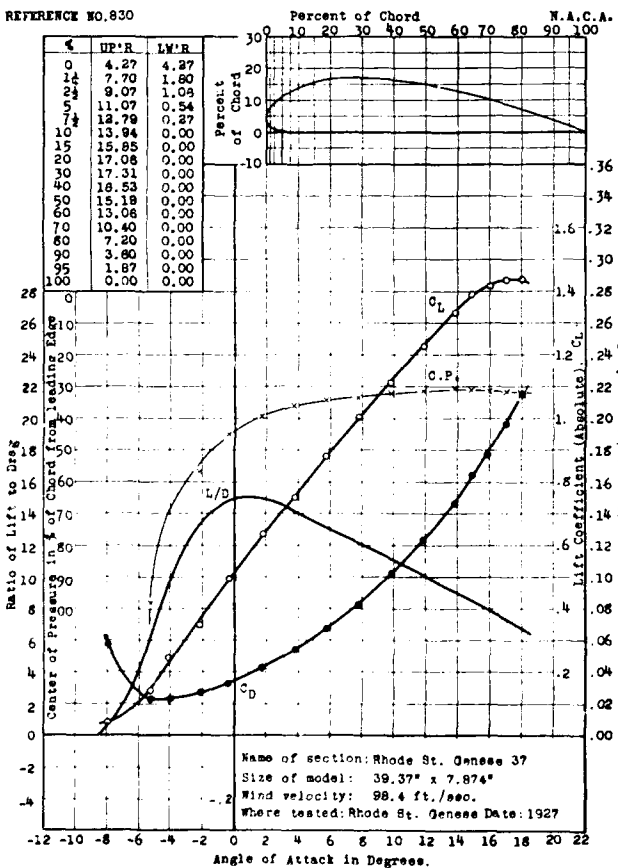
REFERENCE NO. 828



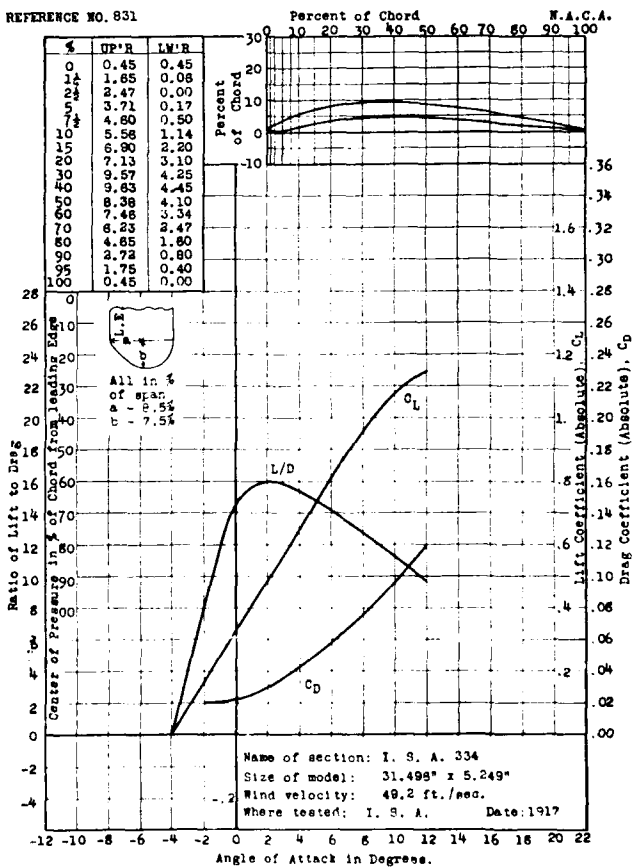
REFERENCE NO. 829



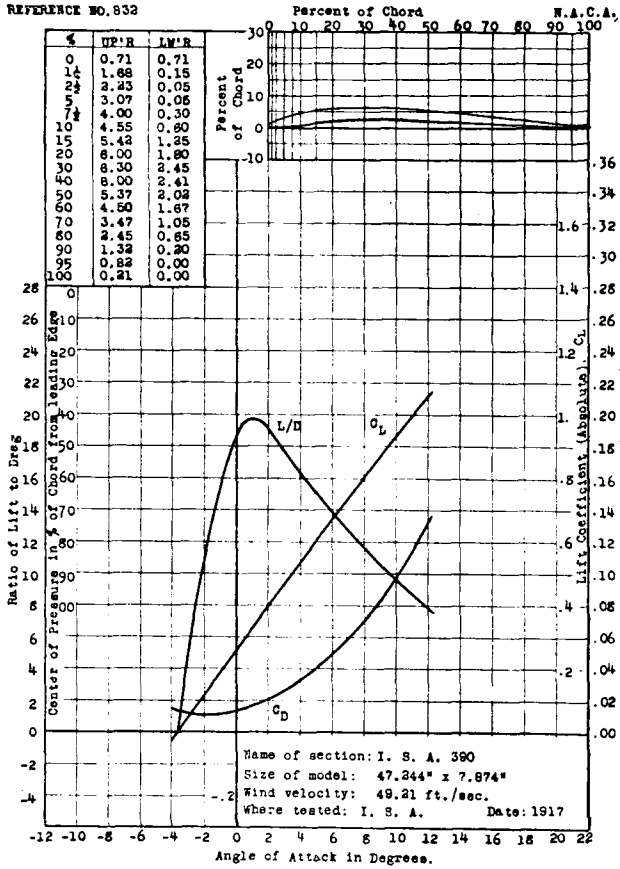
REFERENCE NO. 830



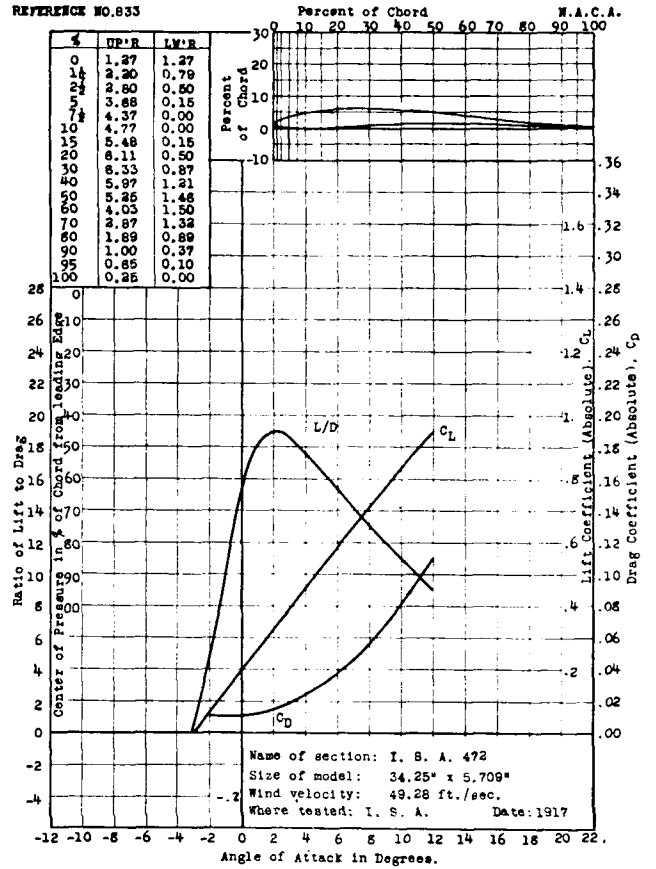
REFERENCE NO. 831



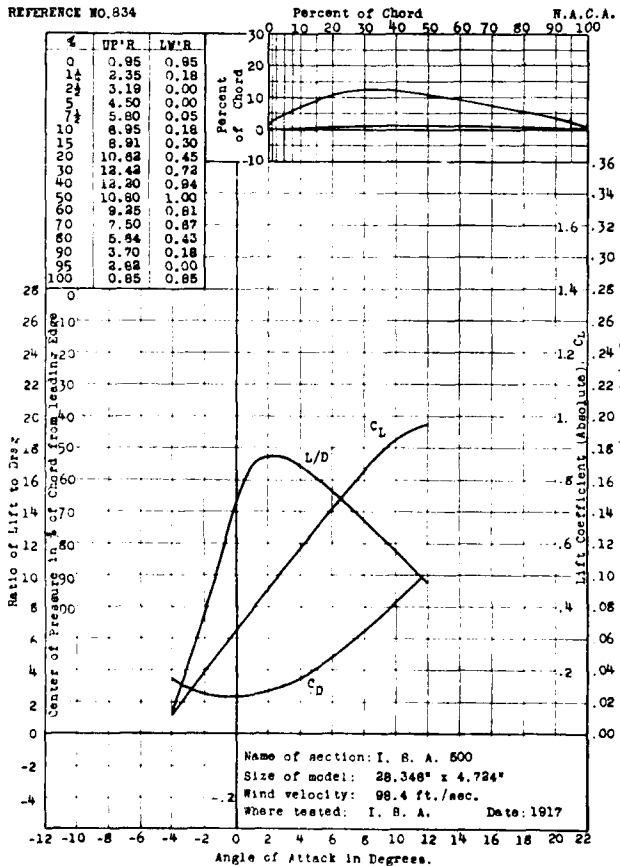
REFERENCE NO. 832



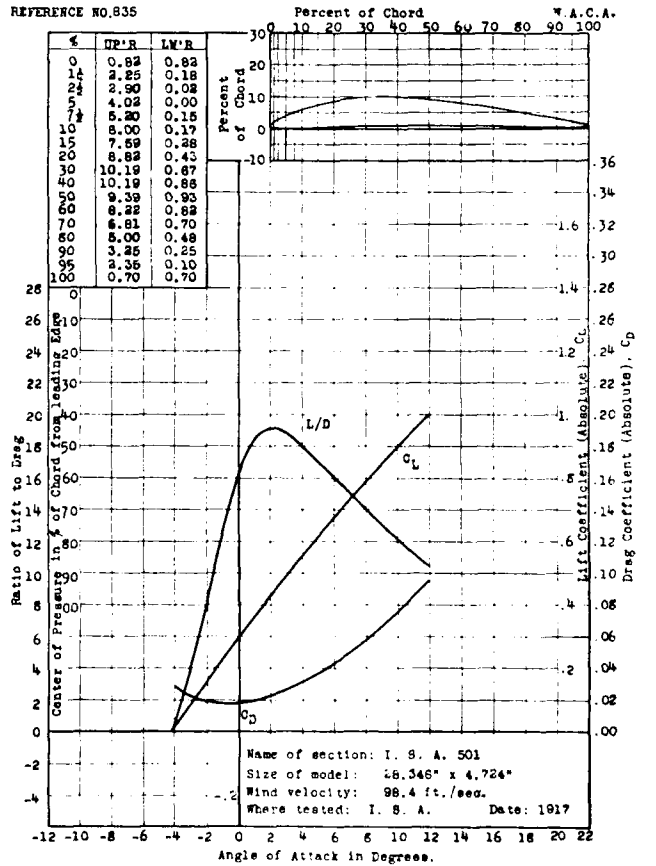
REFERENCE NO. 833

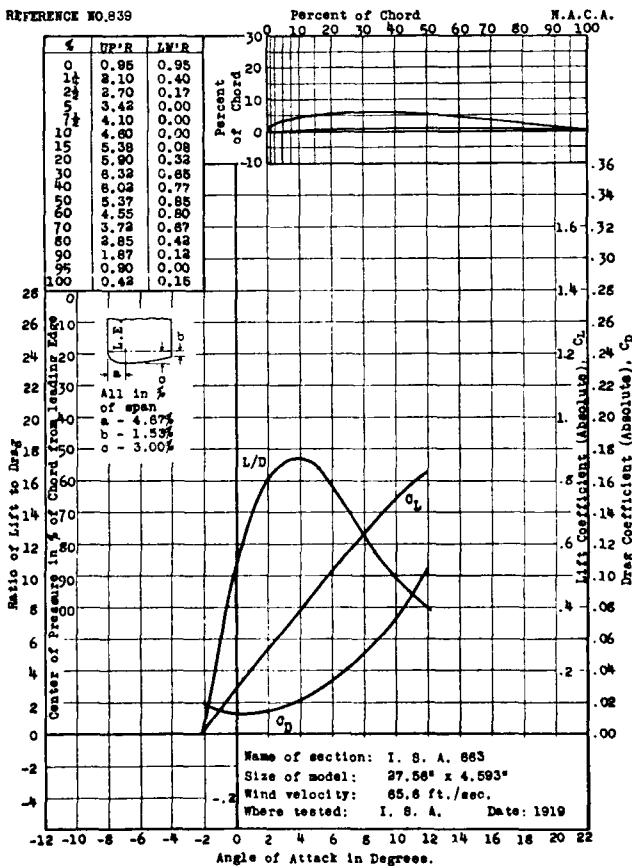
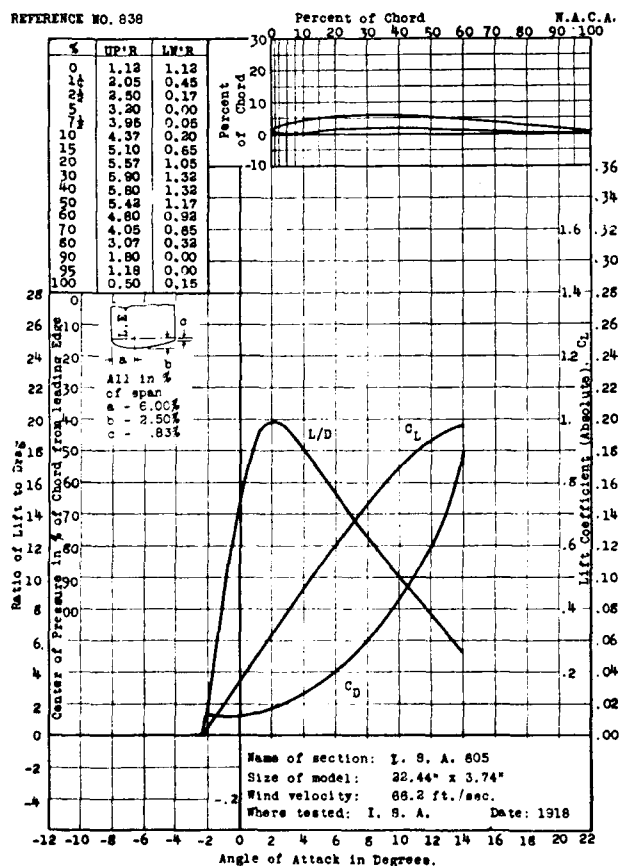
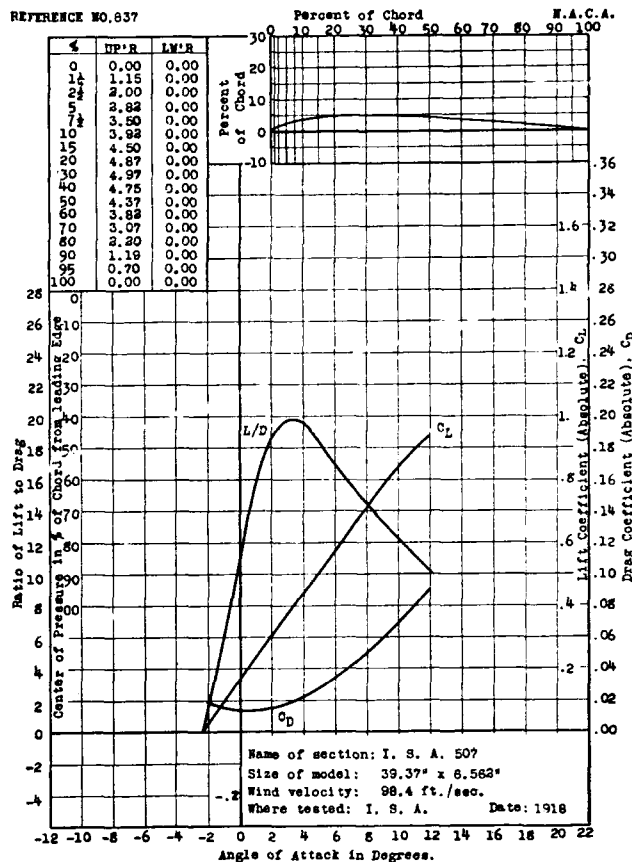
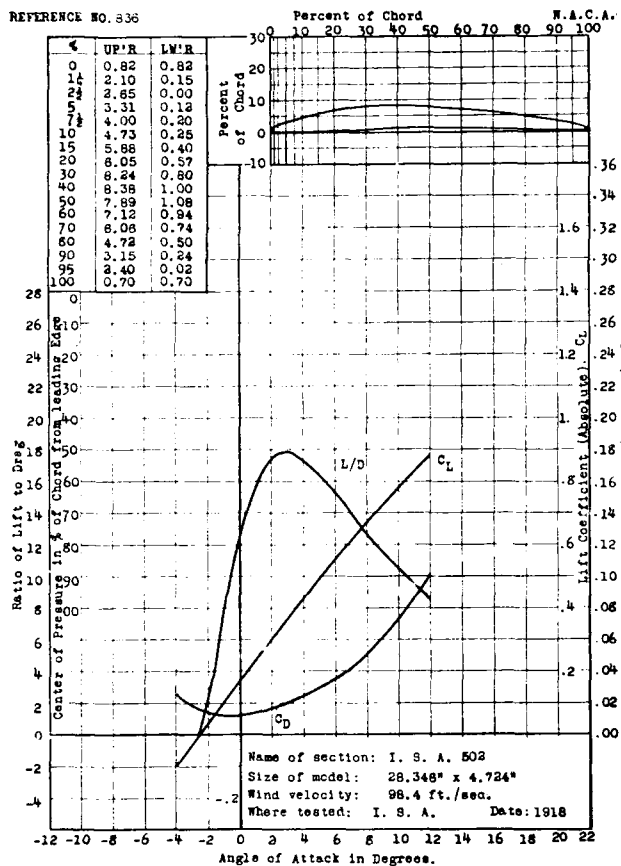


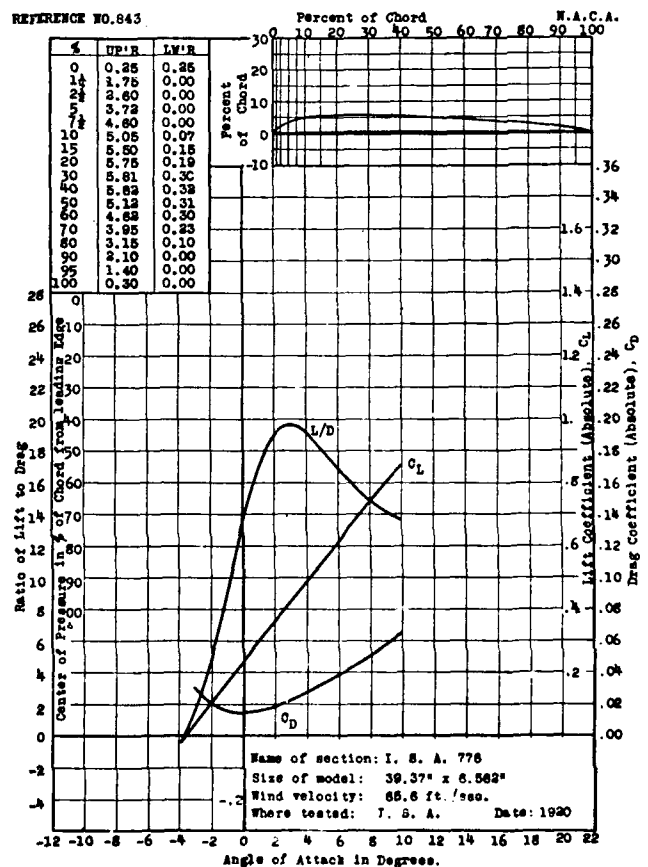
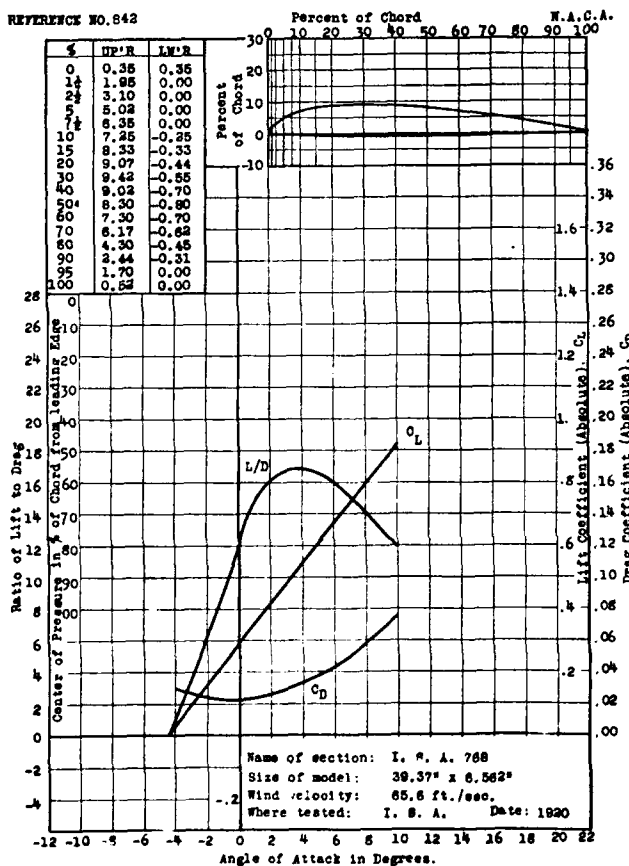
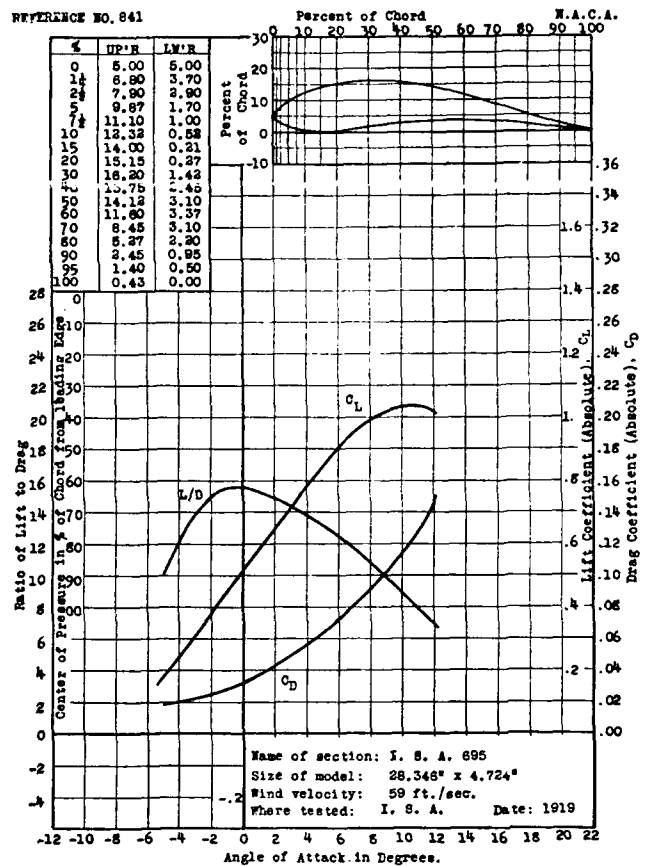
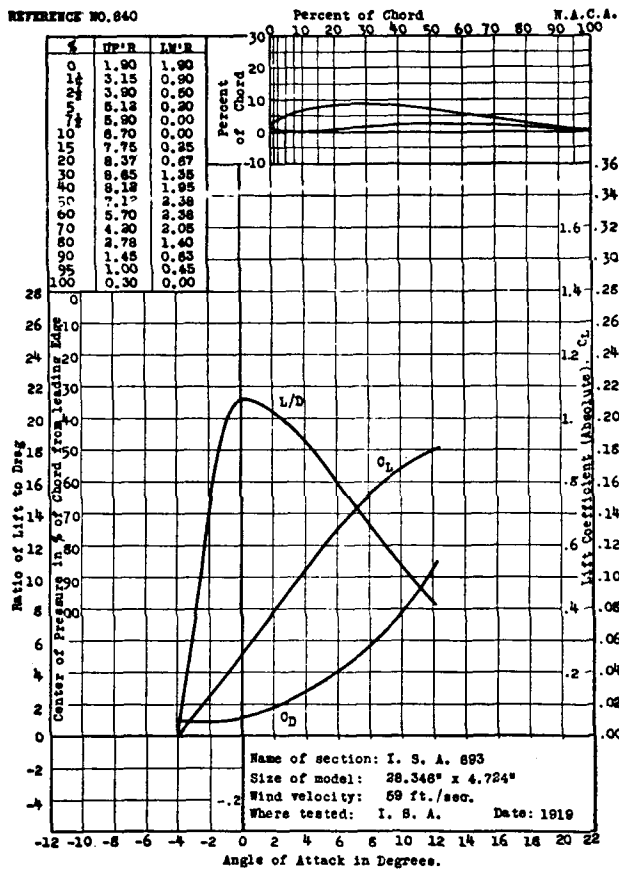
REFERENCE NO. 834



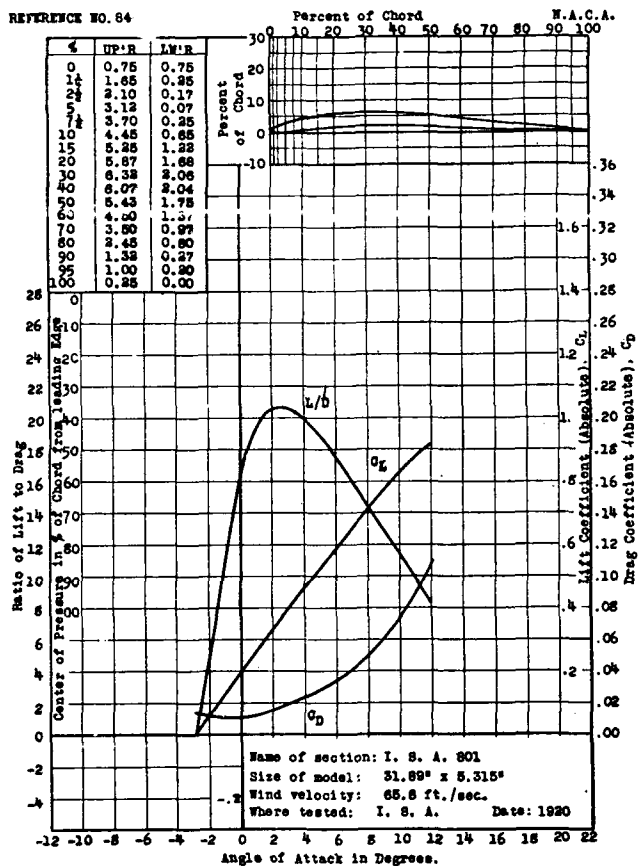
REFERENCE NO. 835



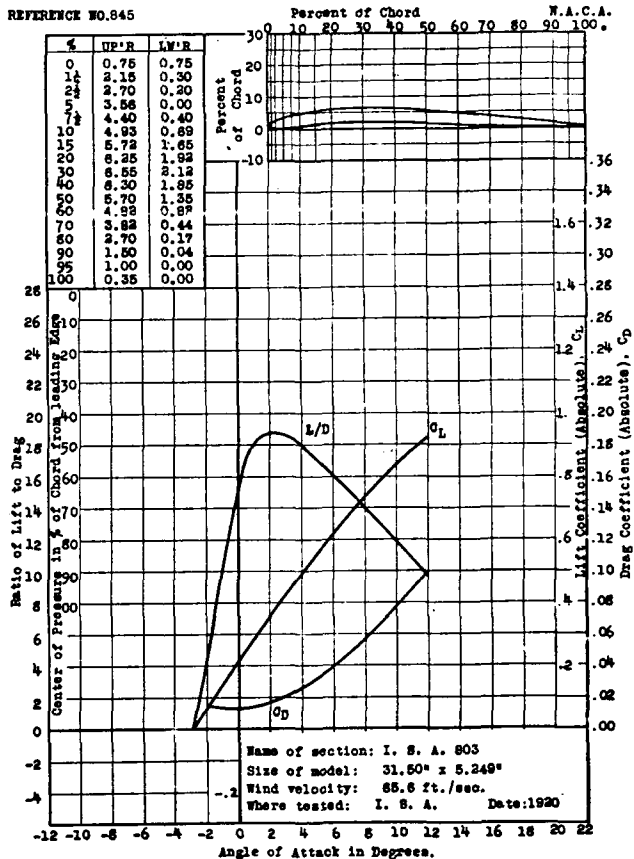




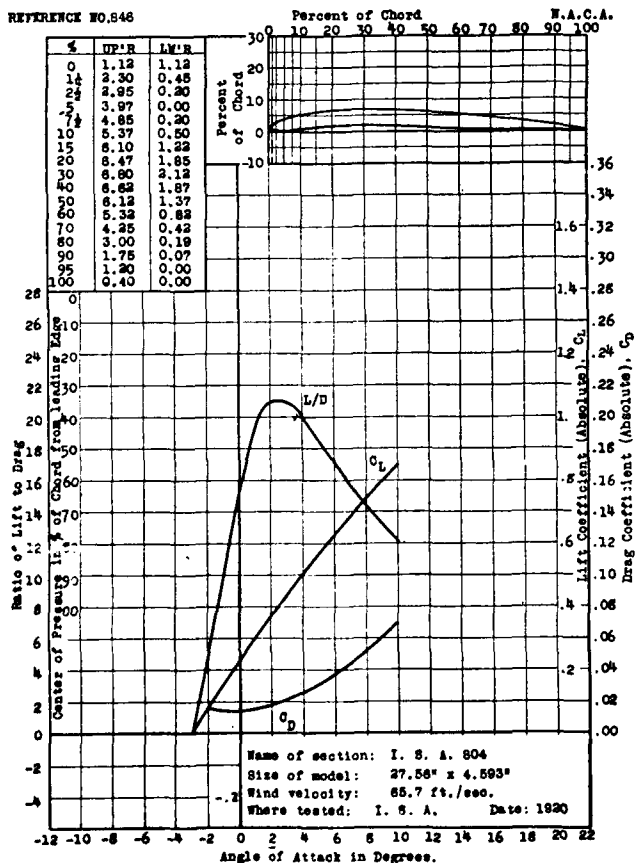
REFERENCE NO.84



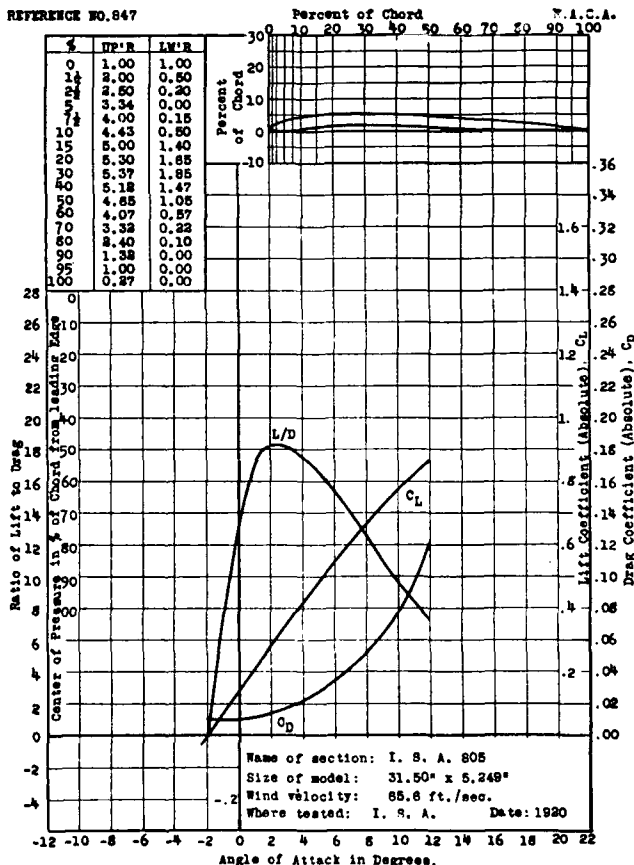
REFERENCE NO.845

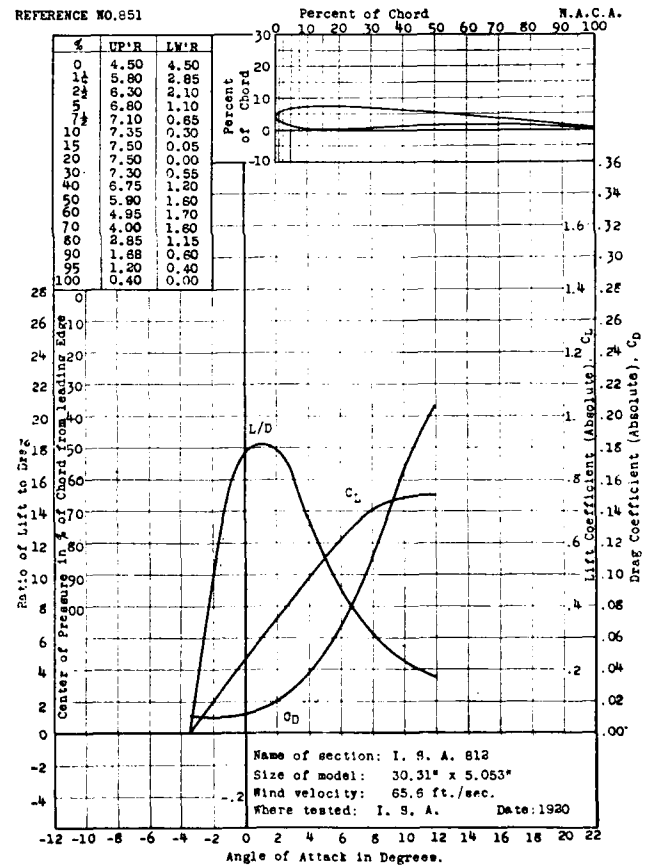
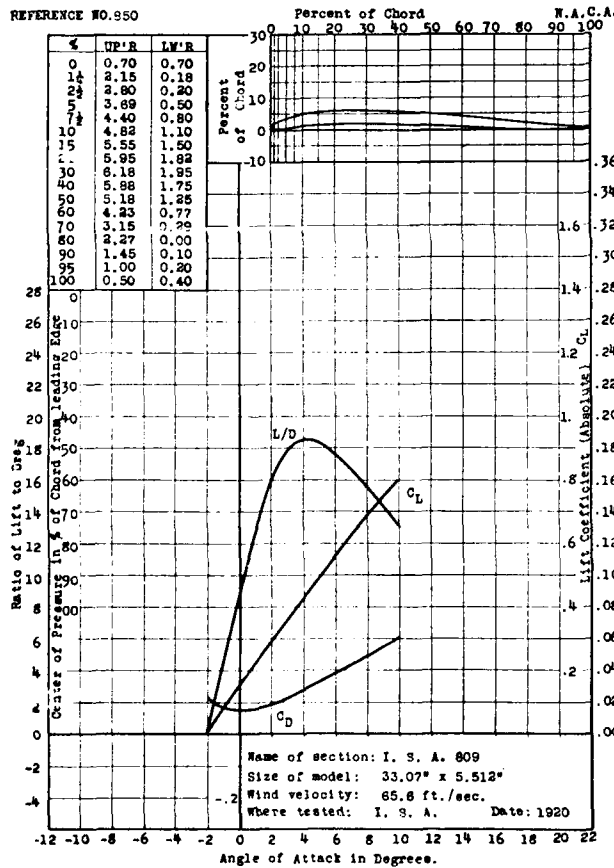
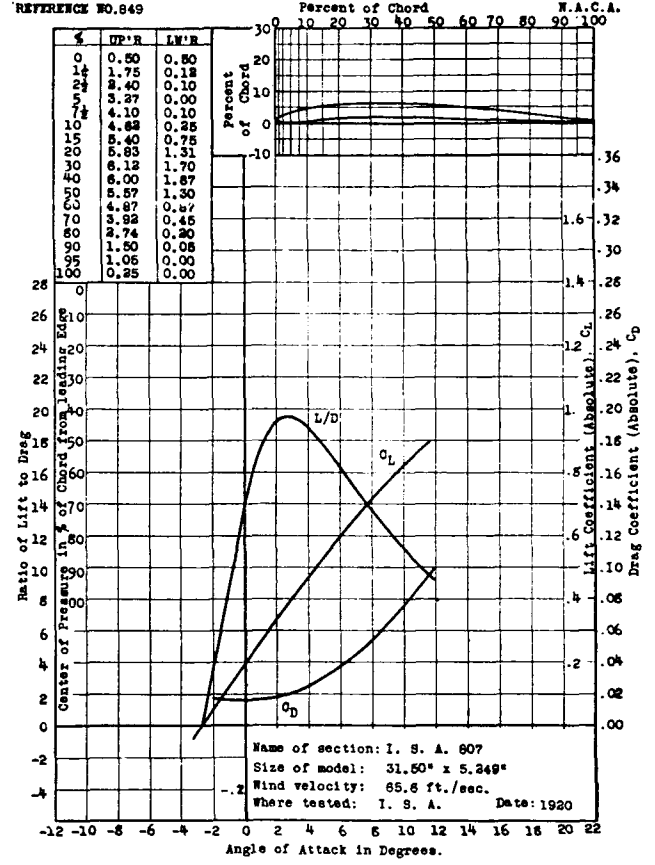
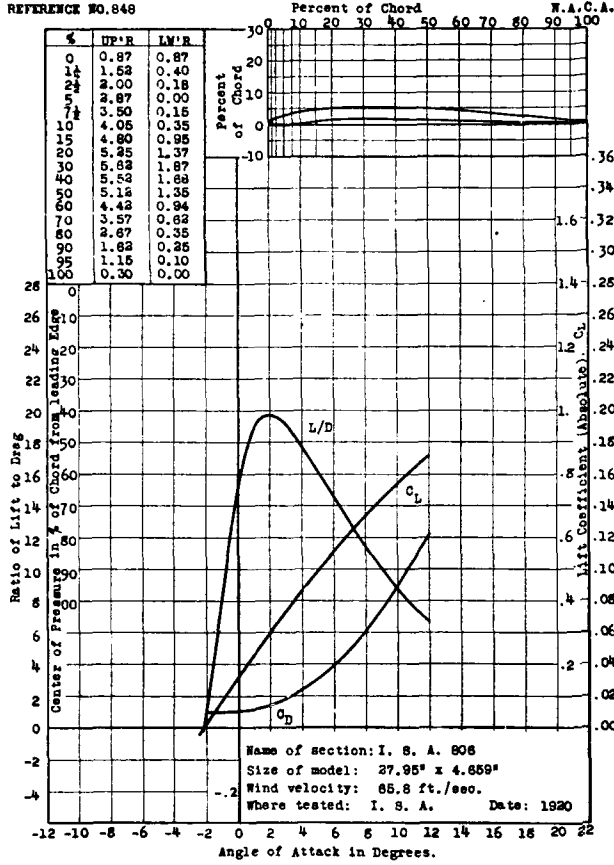


REFERENCE NO.846

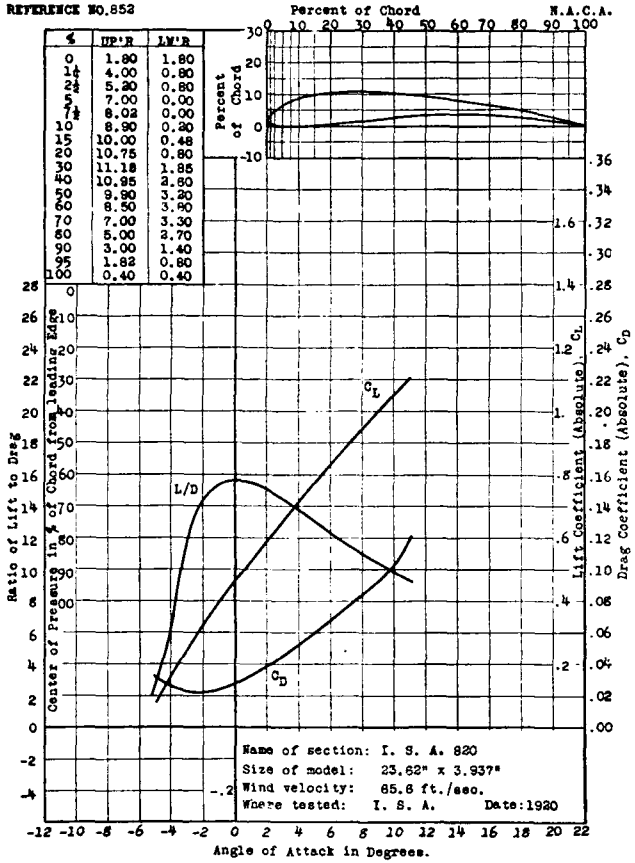


REFERENCE NO.847

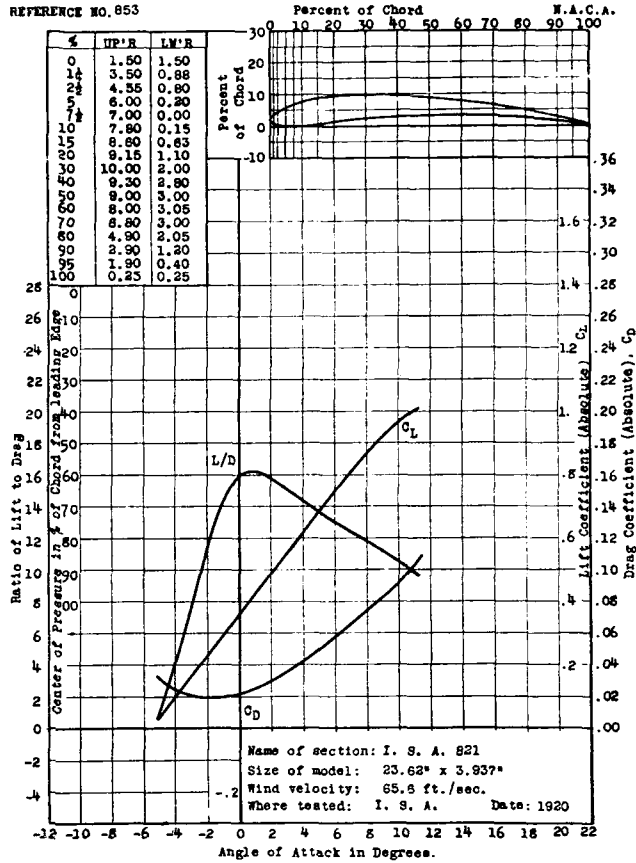




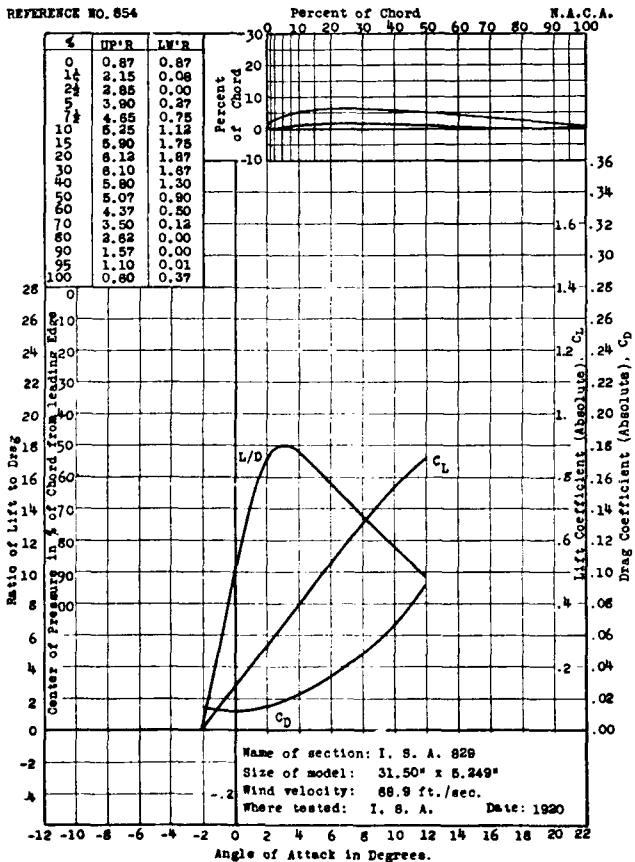
REFERENCE NO. 852



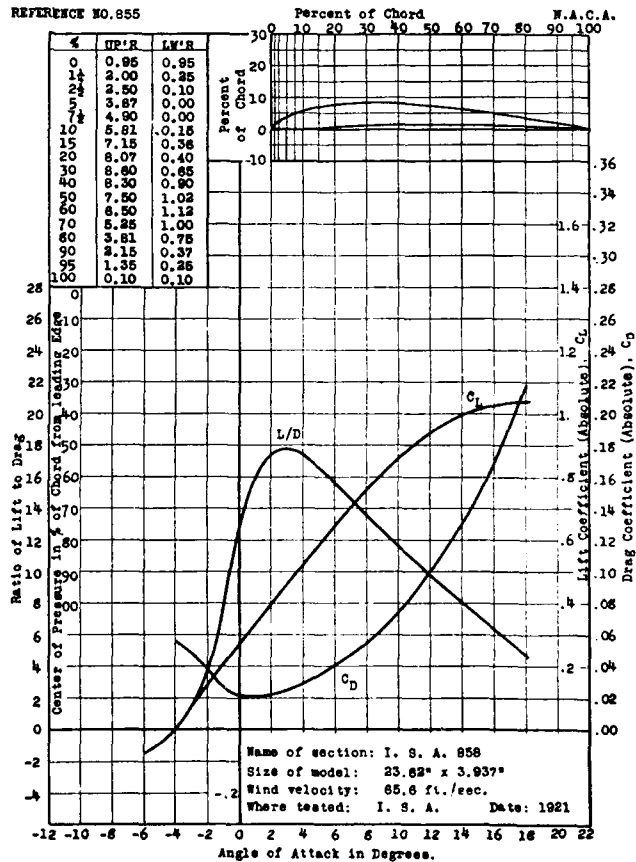
REFERENCE NO. 853

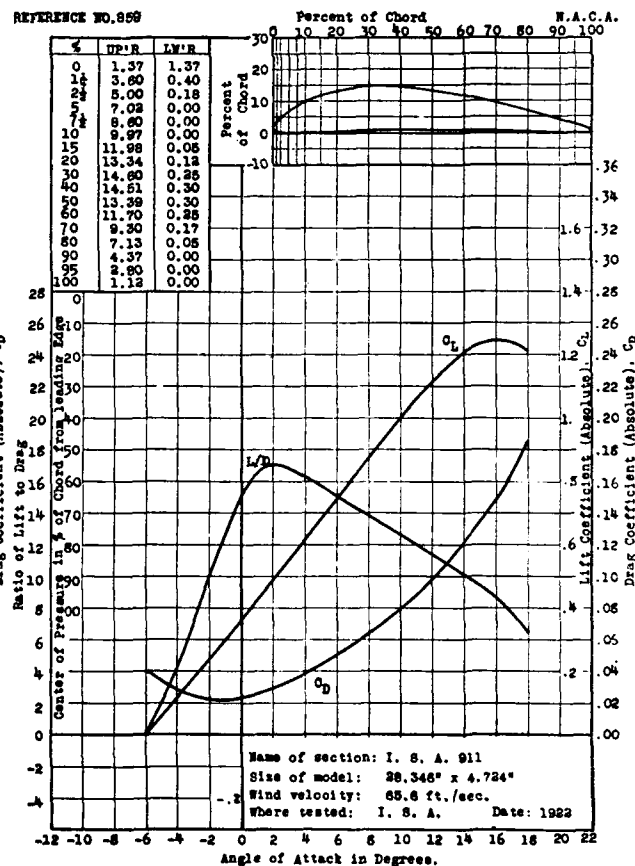
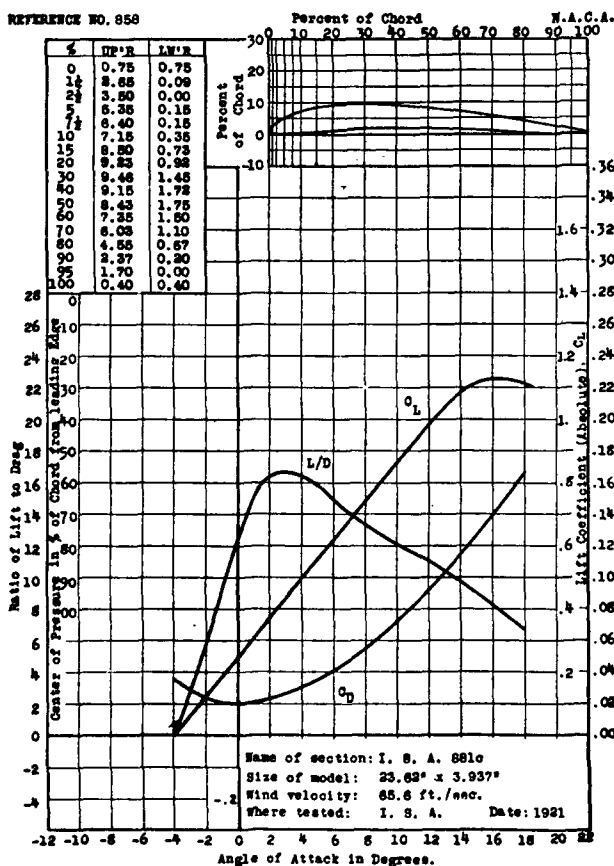
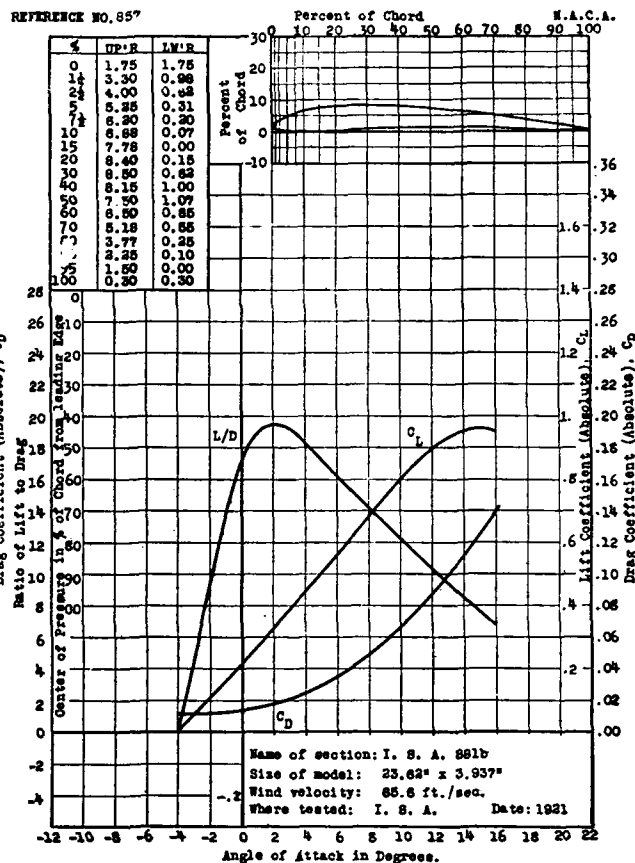
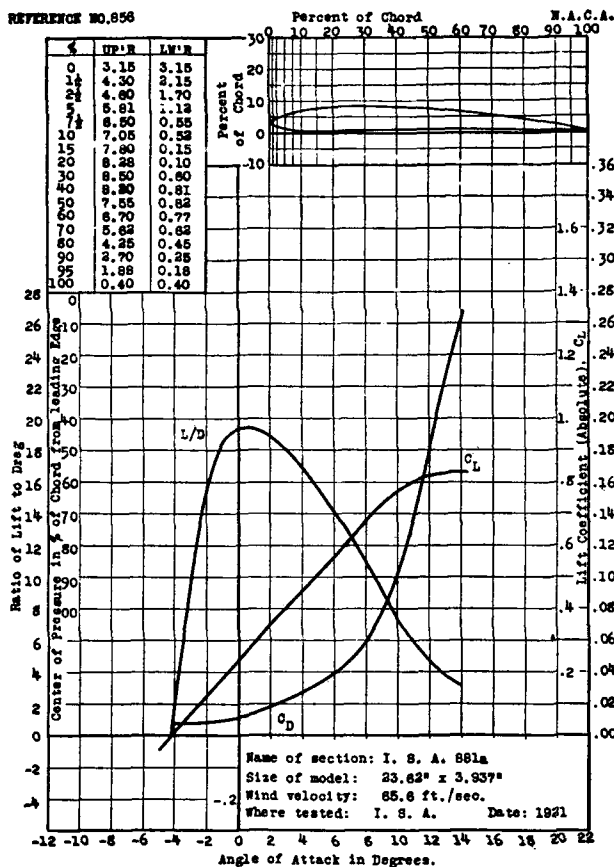


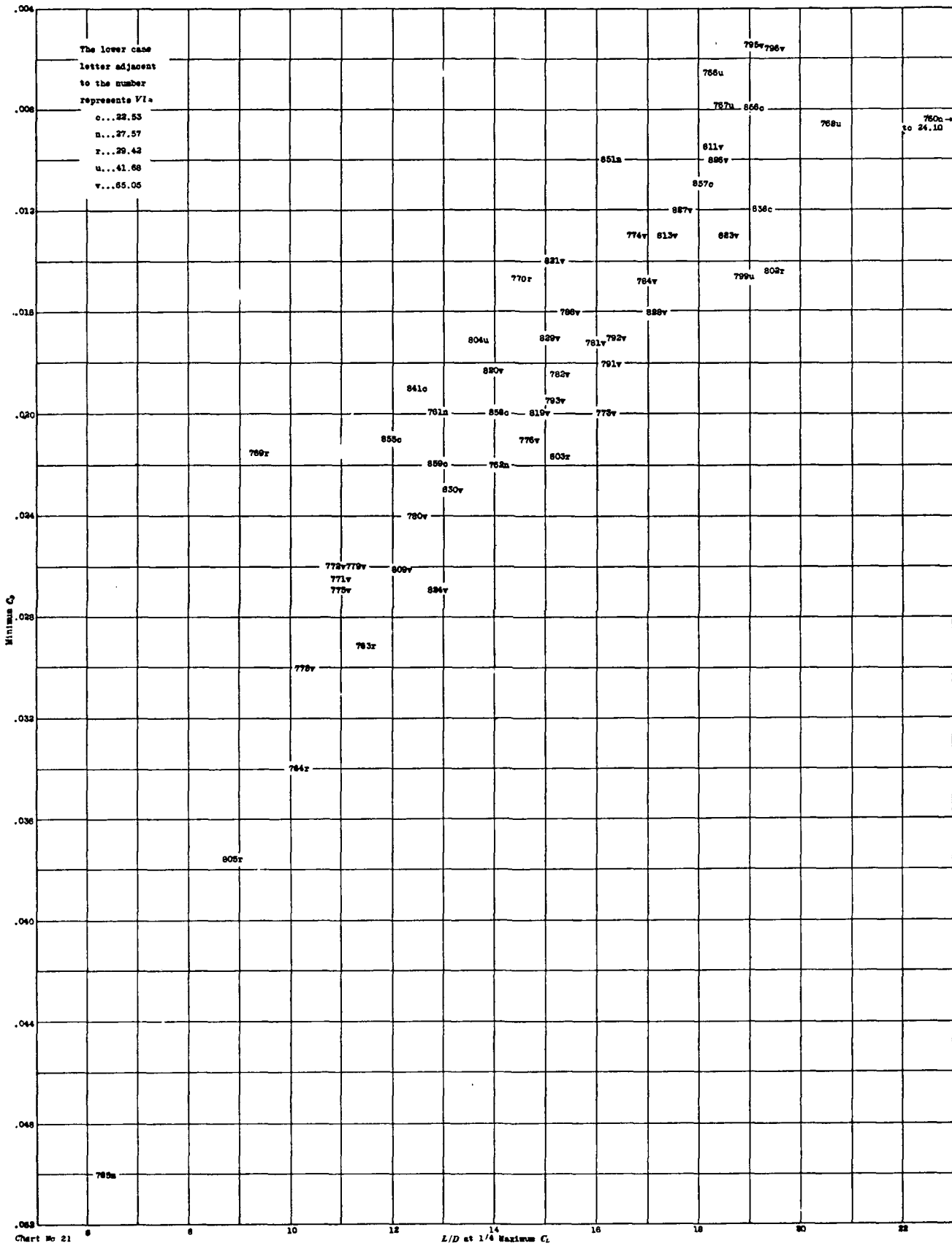
REFERENCE NO. 854

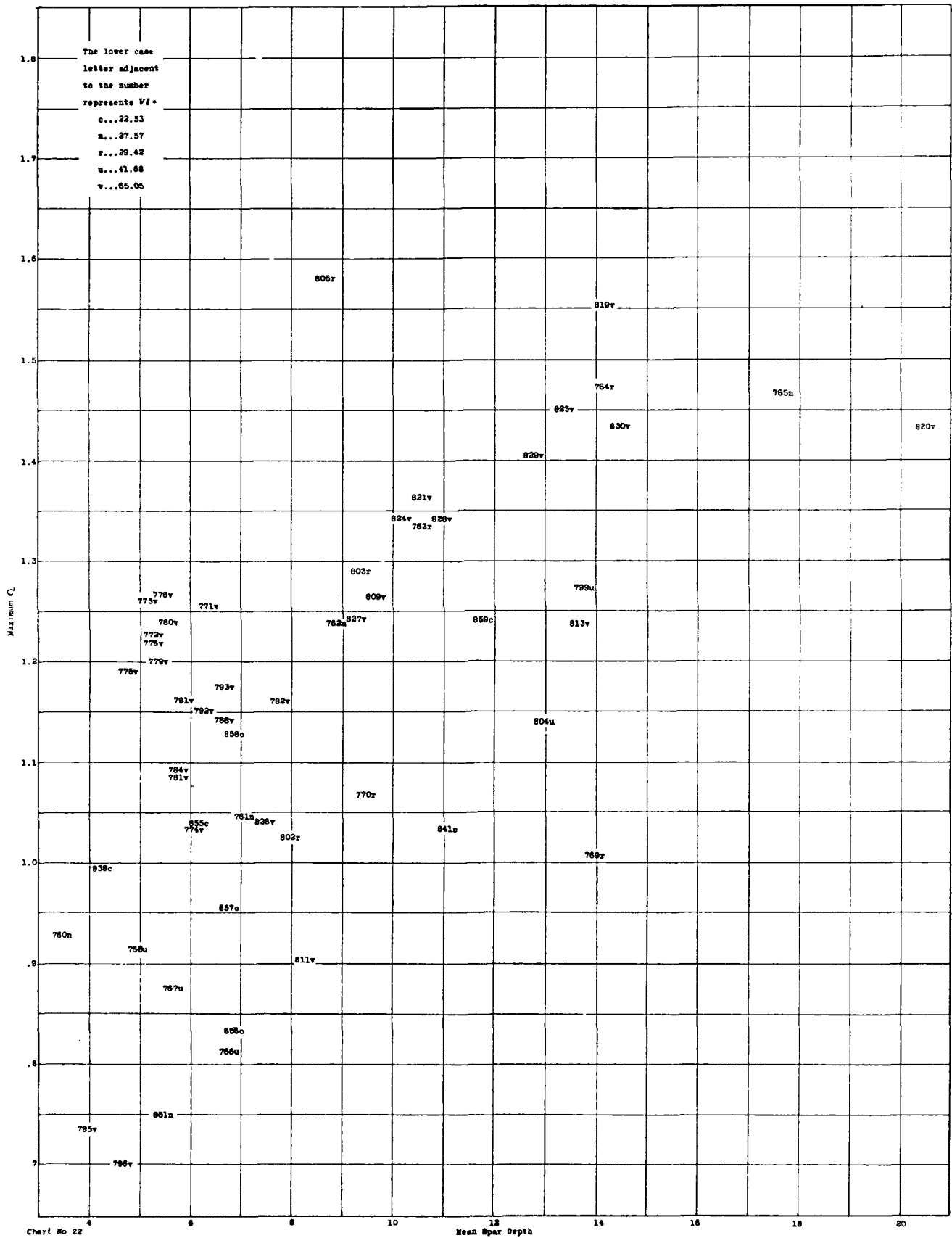


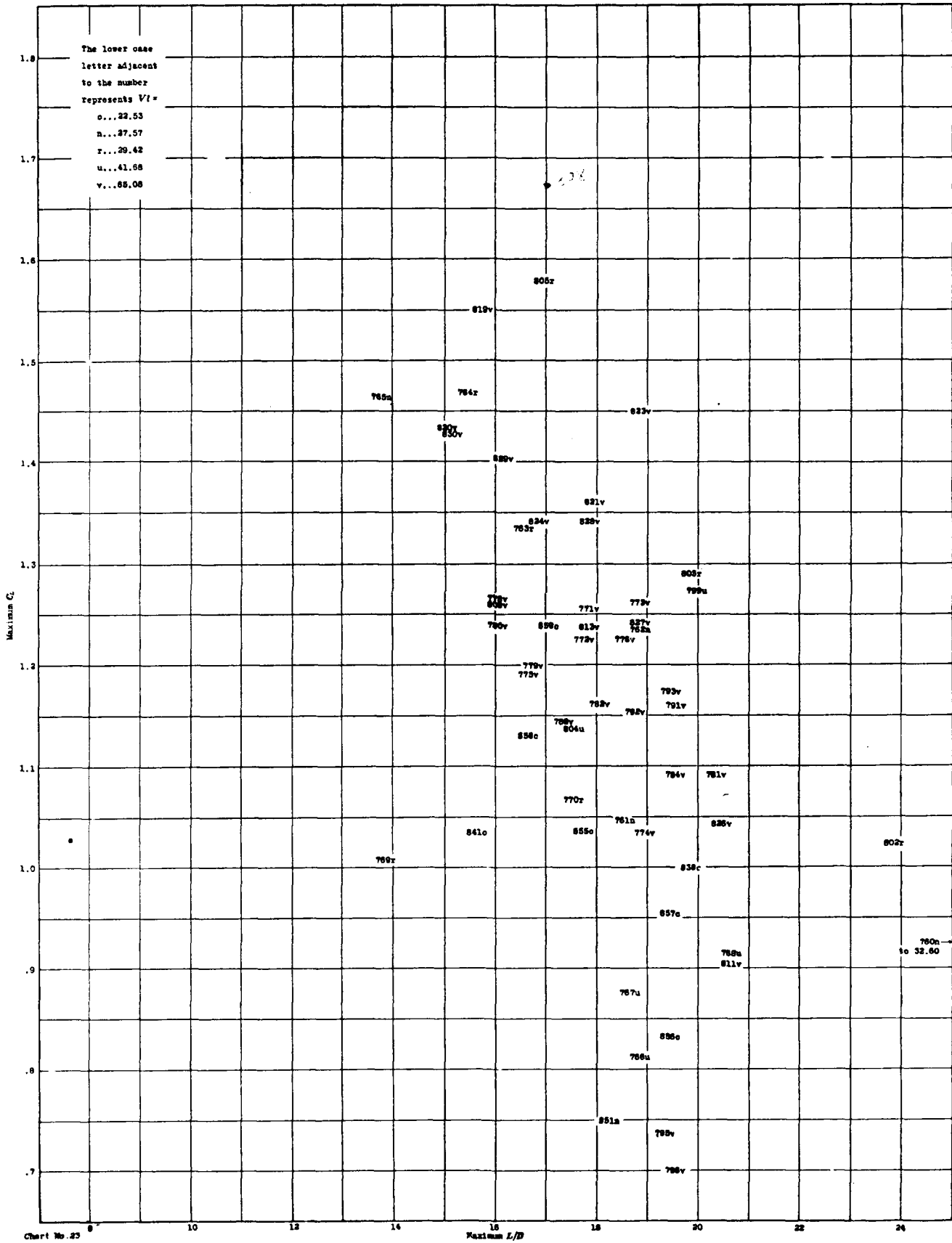
REFERENCE NO. 855

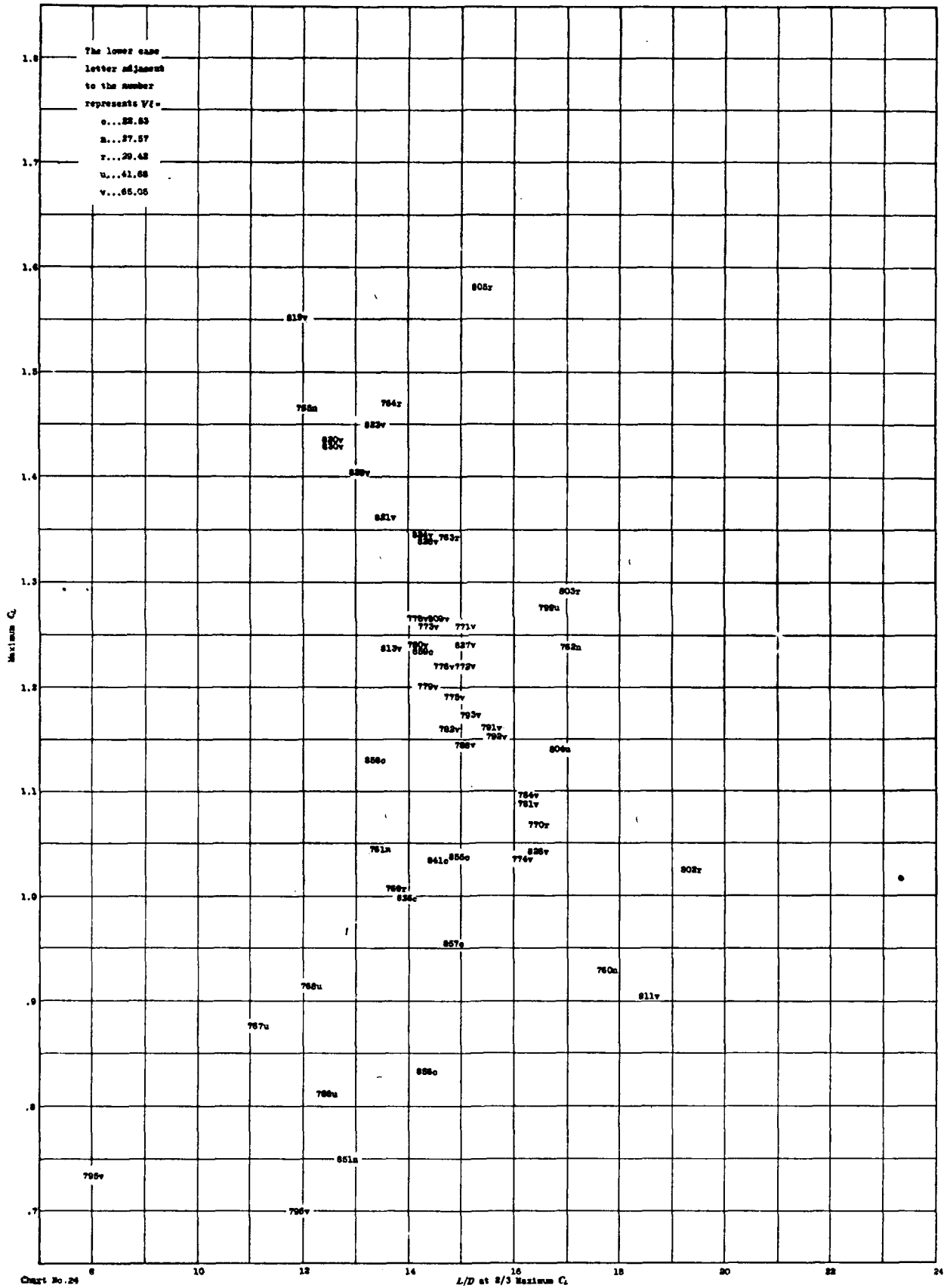




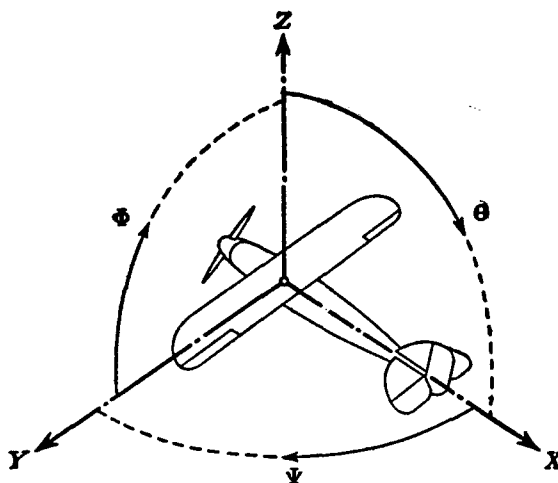








ADDITIONAL COPIES
OF THIS PUBLICATION MAY BE PROCURED FROM
THE SUPERINTENDENT OF DOCUMENTS
U.S. GOVERNMENT PRINTING OFFICE
WASHINGTON, D. C.
AT
20 CENTS PER COPY



Positive directions of axes and angles (forces and moments) are shown by arrows

Axis		Force (parallel to axis) symbol	Moment about axis			Angle		Velocities	
Designation	Sym- bol		Designa- tion	Sym- bol	Positive direction	Designa- tion	Sym- bol	Linear (compo- nent along axis)	Angular
Longitudinal.....	X	X	rolling.....	L	Y → Z	roll.....	Φ	u	p
Lateral.....	Y	Y	pitching.....	M	Z → X	pitch.....	Θ	v	q
Normal.....	Z	Z	yawing.....	N	X → Y	yaw.....	Ψ	w	r

Absolute coefficients of moment

$$C_L = \frac{L}{qbS} \quad C_M = \frac{M}{qcS} \quad C_N = \frac{N}{qfS}$$

Angle of set of control surface (relative to neu-
tral position), δ . (Indicate surface by proper
subscript.)

4. PROPELLER SYMBOLS

D , Diameter.
 p_e , Effective pitch
 p_o , Mean geometric pitch.
 p_s , Standard pitch.
 p_v , Zero thrust.
 p_a , Zero torque.
 p/D , Pitch ratio.
 V' , Inflow velocity.
 V_s , Slip stream velocity.

T , Thrust.
 Q , Torque.
 P , Power.

(If "coefficients" are introduced all
units used must be consistent.)

η , Efficiency = $T V/P$.
 n , Revolutions per sec., r. p. s.
 N , Revolutions per minute., R. P. M.
 Φ , Effective helix angle = $\tan^{-1} \left(\frac{V}{2\pi r n} \right)$

5. NUMERICAL RELATIONS

1 HP = 76.04 kg/m/sec. = 550 lb./ft./sec.
 1 kg/m/sec. = 0.01315 HP.
 1 mi./hr. = 0.44704 m/sec.
 1 m/sec. = 2.23693 mi./hr.

1 lb. = 0.4535924277 kg.
 1 kg = 2.2046224 lb.
 1 mi. = 1609.35 m = 5280 ft.
 1 m = 3.2808333 ft.