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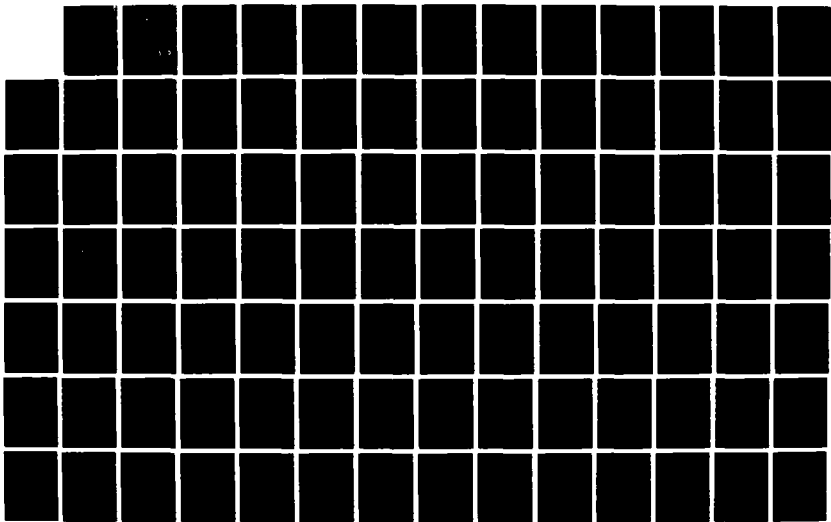
F-14 WING OUTER PANEL FATIGUE TEST SPECTRUM(U) NAVAL  
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F-14 WING OUTER PANEL  
FATIGUE TEST SPECTRUM

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Warminster, Pennsylvania 18974-5000

APRIL 1987

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## SYMBOLS

B.L.	Butt Line (= y)
C.G.	Center of Gravity
E	Exceedances
F.S.	Fuselage Station (= x)
$F_x$	Horizontal Shear ~ Pounds
$F_z$	Vertical Shear ~ Pounds
M	Mach No.
$M_x$	Moment About x Axis (unswept) ~ Inch-Pounds
$M_y$	Moment About y Axis (unswept) ~ Inch -Pounds
$M_z$	Moment About z Axis (unswept) ~ Inch-Pounds
$M_x'$	Moment About x' Axis (WOP axis) ~ Inch-Pounds
$M_y'$	Moment About y' Axis (WOP axis) ~ Inch-Pounds
$N_x$	Longitudinal Load Factor ~ g's
$N_y$	Lateral Load Factor ~ g's
$N_z$	Normal Load Factor ~ g's
$\dot{\phi}$	Roll Acceleration ~ Radians/sec <sup>2</sup>
q	Dynamic Pressure ~ Pounds Per Square Foot
W.L.	Water Line (= z)
$V_{wa}$	Sinking Speed ~ Feet Per Second
$\alpha$	Wing Leading Edge Sweep Angle ~ degrees
$\Delta\alpha$	Change in Sweep Angle ~ Degrees
$V_\infty$	Free Stream Velocity ~ Feet Per Second
$\alpha$	Angle of Attack ~ Degrees

SUMMARY

A spectrum of loads for the F-14 Wing Outer Panel fatigue test has been derived, based on the spectrum used by Grumman for the Aircraft #98 test of the F-14A fuselage. Flight and ground load distributions on the wing have been developed corresponding to those in that test, and the Aircraft #98 spectrum was simplified where significant fuselage loading conditions would have negligible effect on the wing outer panel. In addition, requirements for sweeping the wing under load have been incorporated based on fleet operational data.

## INTRODUCTION

A fatigue test is planned, under Naval Air Systems Command (NAVAIR) sponsorship, for the F-14 Wing Outer Panel, including the wing pivot and the structure of the variable sweep panel outboard of the pivot. This part of the F-14 airframe had not been included in the Aircraft #98 fatigue test conducted by Grumman, and so is the remaining structure to be demonstrated to full service lifetime under a realistic flight-by-flight block loading.

## AIRCRAFT #98 FATIGUE TEST CONDITIONS

The so-called Aircraft #98 Fatigue Test was conducted by Grumman under NAVAIR Contract beginning in 1977. The purpose of the test was to demonstrate the structural integrity of the fatigue-critical portions of the F-14A airframe, such as the aft fuselage, under a realistic spectrum of loads. The original F-14A fatigue test had undergone many structural beef-ups and spectrum changes in mid-test, so that a re-test to a more complete spectrum representative of actual service operation was required. In addition, modern testing philosophies such as flight-by-flight block loading were incorporated in the Aircraft #98 test. The test article was the fuselage of F-14A shop No. 98, (BUNO 159432) which had sustained damage in a ground accident. The fuselage was repaired and brought to current ECP status by Grumman. The wings were considered less fatigue-critical than the fuselage, and were not included in the test.

It should be noted that since the wing was not included, test loadings were selected based on their criticality to the fuselage. For any given loading condition in the spectrum, the aspects of that condition significant to the fuselage structure were included. Also, only the loads at the wing pivot - shear, moment and torsion - were needed for the Aircraft #98 test; wing outer panel load distributions were not constructed.

Spectrum

The fatigue test spectrum for the Aircraft #98 fatigue test was described in references (a) through (c). The test was intended to demonstrate a fatigue life of 12000 hours. Grumman developed their spectrum in terms of a 480 hour block, a repeatable flight-by-flight sequence with a selected wing sweep schedule. Each 480 hour block consisted of 440 flights. Flight loads consisted of symmetric maneuvers, level rolls, and turn reversals (rolling maneuvers at 3.5 g or greater). The spectrum of flight maneuvers, derived in part from the F-14A Fleet Flight Loads Survey as reported in reference (d), included a breakdown into five wing sweep positions, and a total of 11528 maneuvers per 480 hour block. Of the 440 flights, 160 included carrier takeoffs (catapults) and arrestments. The remaining 280 were field landings and field carrier landing practice (FCLP). Touch-and-go's were included in both carrier and field landings.

Table I, extracted from reference (e), lists the 378 load conditions that made up the Aircraft #98 spectrum. The actual sequential spectrum for a 480 hour block is included in Appendix (A). This spectrum is the one finally applied in the Aircraft #98 test, after some of the lower level loads in the original spectrum were truncated out.

Flight Loads

Derivation of the balanced loads for flight conditions of the aircraft #98 test was documented in reference (b). The contractor measured net loads on a flight test aircraft, extracted airloads from these net loads so as to construct a rational aero model, and then used the aero model and known inertias to develop balanced loads.

A symmetric maneuver load cycle consisted of a peak  $N_z$  of 3.5g or greater followed by a valley  $N_z$  of less than 1.0g. Wing loads in symmetric maneuvers were found to peak at about  $8^\circ$  angle of attack. At higher  $N_z$ 's the additional lift was generated by the fuselage. Therefore, for each wing sweep angle, the peak wing load occurred at a different  $N_z$  which was generally lower than the maximum  $N_z$  in the spectrum. The aircraft #98 fatigue test load sequence included this peak in the increasing  $N_z$  portion of the cycle. A complete symmetric maneuver sequence was as shown below:

1.0g →  $N_z$  for → Maneuver →  $N_z$  → 1.0g  
 peak wing peak valley  
 load

An asymmetric maneuver load cycle consisted of a symmetric  $N_z$  condition with an antisymmetric (roll acceleration) component superimposed in each direction sequentially. For a turn reversal, which is an asymmetric maneuver at 3.5g or greater, this sequence was as shown below:

1.0g → Maneuver  $N_z$  → Maneuver  $N_z$  → Maneuver  $N_z$  →  $N_z$  → 1.0g  
 +p -p valley

For level rolls, the maneuver  $N_z$  was 1.0g. There were no  $N_z$  valleys associated with level rolls.

Ground Loads

Derivation of the balanced loads for ground conditions of the Aircraft #98 test was documented in reference (c). The contractor conducted an extensive series of transient analyses and correlated his model with the results of full scale drop tests and instrumented aircraft landing tests. In the final load sequence, each landing was a function of sinking speed and landing attitude and included up to four cycles of load.

For an arrested landing, the longitudinal loads of the arrestment were considered to follow separately after the vertical loads of the landing.

For a catapult, a sequence of loads including buffing, release, and asymmetric tow loads was applied.

## DERIVATION OF WING OUTER PANEL TEST CONDITIONS

The spectrum for the F-14A Wing Outer Panel fatigue test has been derived primarily from the Aircraft #98 test spectrum. In addition, this test is to evaluate the effects of wing sweeping, and so provisions for sweeping the wing under load have been added. Since the Aircraft #98 test wing loads consisted of net loads at the pivot only, a set of distributions has been developed using theoretical aerodynamic analysis. Finally, the Aircraft #98 spectrum has been shortened, primarily in the ground loads area, by simplifying the loading conditions.

The sign conventions used throughout this report are shown in Figure 1. Aircraft #98 test condition loads were derived in an unswept coordinate system for application to the fuselage. However, for this test of the Wing Outer Panel, a coordinate system that remains attached to the wing as it sweeps is useful. Therefore, the wing quarter chord (approximately) line, which passes through the pivot and is at an angle of  $-3.5^\circ$  to the leading edge line, is used. This axis system is shown in Figure 1c.

Loads for each test condition consist of pivot location resultants shown in Table II, with distributions that can be constructed from component distributions provided in various other Figures. Wing pivot resultants have five components; no lateral loads are included.

Flight Loads

Derivation of the wing pivot loads for symmetric flight conditions followed the procedures used by Grumman. Table III contains these calculations, and the resulting net pivot loads are considered identical to those that were applied in the Aircraft #98 test. The values of shear, bending moment and torsional moment about unswept axes through the pivot are plotted in Figures 2 through 6 for the various sweep angles. The values of these loads at  $N_2$ 's other than those calculated were obtained by linear interpolation.

In order to determine the distribution of the airloads over the wing outer panel, a theoretical airloads analysis was performed using the RAPIDLOADS computer code, reference (f). The aerodynamic model of the wing in the  $25^\circ$  sweep position is shown on Figure 7. The model included the effects of camber, twist, dihedral, and fuselage interaction. As with Grumman's Aircraft #98 conditions, no maneuvering flaps or slats were used. The flight conditions for  $25^\circ$  sweep were at Mach 0.62, 10,000 feet altitude. The agreement of this analysis with Grumman's Aircraft #98 data, for the linear region up to an angle of attack  $\alpha = 3.2^\circ$ , can be seen in Figure 3.

Net loads were obtained by combining these airloads with inertia loads. Inertia loads were obtained by constructing a distribution of the 2019 lb empty weight of the wing (for flight maneuvers at the flight design gross weight, there is 90% internal fuel, which is all in the fuselage). For the maximum load condition at  $25^\circ$  sweep, the net loads are as shown in Figure 3.

The loads shown in Tables II and III are in the unswept axis system and in the wing outer panel axis system illustrated in Figures 1b and 1c, respectively. The net loads shown in Figure 9 are located along the Wing Outer Panel quarter chord (Y' axis) illustrated in Figure 1c, rotated into bending and torsion of the quarter chord ( $M_x'$  and  $M_y'$  respectively). This figure, therefore, represents a distribution in suitable format for test implementation.

In order to avoid the necessity to derive individual distributions for each flight maneuver condition (which would also complicate the test), an assessment was made of the similarity of these distributions for different  $N_z$  and sweep values using Grumman's flight data. Aero data from reference (5) was used to determine the variability of the center of pressure in the axis system that moves with the wing. This variability is shown in Table IV, and illustrated on Figure 10. Inertia load, the other component of the net load distribution, is of course invariant with wing sweep. Therefore, the variation was concluded to be small enough to justify using the same spanwise load distribution for all symmetric flight maneuver load conditions. Scaling the spanwise load distribution will not necessarily match the root torsion value; however, the torsion distribution of Figure 9 should be scaled separately to match the root value.

In the Aircraft #98 test, asymmetric maneuver loads were made up of a symmetric load component, as previously derived, and an antisymmetric component due to rolling acceleration. Since the F-14 uses spoilers and differential tail for roll control instead of ailerons, the additional load in a roll acceleration was considered to come from inertia only. The distributed inertia load due to a unit roll acceleration is shown in Figure 11. This distribution is applicable to both level (1 g) rolls and turn reversals, and can be added to the symmetric load distribution.

#### Ground Loads

During landing, the wing experiences a reversed loading as it goes from an upward airload to a downward inertia load. The relationship between sinking speed and  $N_z$  shown in Figure 12 was used. The wing is considered to be empty of fuel and in the fully forward  $20^\circ$  sweep position. Wing weight is 2019 lb, from reference (3). The unit distribution of inertia load is shown in Figure 13. While the aircraft #98 test used a number of load applications per landing, and a distribution of landing attitudes, for the Wing Outer Panel test only one cycle per landing was assumed, and the acceleration was taken as a function of sinking speed only.

Load factors for arrestment conditions were given in reference (6), and are reproduced here in Table V. The wing was considered to be empty of fuel and positioned at the forward  $20^\circ$  sweep position, the same as for the landing conditions. The inertia load distribution of Figure 13 can be scaled to the proper  $N_z$  value and applied in the longitudinal direction.

Catapult conditions were also described in reference (c) and are reproduced here in Table VI. For catapult, the wing was considered to be full of fuel, 2108 lbs per wing, reference (3), and at the 20° sweep position. The unit distribution of inertia load for the full fuel condition is shown in Figure 14. While the Aircraft #98 test used a series of spotting, buffing, tow and release loads to represent a catapult, just one load application per catapult was used for the Wing Outer Panel test.

### Spectrum

The Aircraft #98 spectrum in Appendix (A) was modified to produce the F-14 Wing Outer Panel spectrum.

First, simplifications applied to the catapult, landing, and arresting load conditions resulted in deleting some conditions and combining others. Then, any sequential duplications of the same load within a flight were removed. The resulting spectrum is given in Appendix (B), and is available on computer tape. The number of applications of each condition in a 480 hour block is shown on Table II, along with the corresponding number from the original spectrum. Since an asymmetric maneuver involves a sequence of conditions including a symmetric maneuver case, the number of actual maneuvers in the spectrum is not readily apparent from Table II. Table VII, however, gives a summary of the maneuvers included in the spectrum, and the normalized total  $N_z$  exceedance curve is plotted in Figure 15.

### Pivoting

The amount of wing sweep movement during normal F-14A flight operations was tabulated by Grumman from F-14A Fleet Flight Loads Survey data. The Grumman data is included herein as Table VIII. It is noted that pivoting from forward to aft positions usually occurs during lightly loaded (accelerating in level flight) conditions, while pivoting from aft to forward positions often occurs in higher g maneuvers, when the drag slows the airplane down.

It was decided, for simplicity with no loss of generality, to impose pivoting under load only on the symmetric flight conditions. Each time a symmetric maneuver ( $N_z = 3.5g$  or greater) is imposed, a wing pivot cycle will be imposed as described in Figure 16. All pivot cycles will be for 10° except those at  $\alpha_{sw} = 25^\circ$  which will be for 5°. With this test procedure, all ramping up or down of load will occur with the wing stationary, and the number of pivoting movements under load will approximate the total shown in Table VIII for  $\alpha > 5^\circ$ .



References:

- (a) GAC Report A51-30-R-80-01 (LD 303-110.4): F-14A Fatigue Criteria for Flight Operations for A/C No. 98 Test Program; 9 Feb 80
- (b) GAC Report A51-30-R-80-03 (LD 303-102.1): F-14A A/C 98 Fatigue Test Program Balanced Aircraft Flight Loads; 15 Feb 80
- (c) GAC Report A51-30-R-80-04: Derivation of Ground Loads for F-14A Aircraft No. 98 Fatigue Test; 15 Apr 80
- (d) GAC Report FAD-303-3-3.31: F-14 Fleet Flight Loads Survey Final Report; 16 Nov 79
- (e) GAC Report SAR 80-3: 30 Apr 80
- (f) NADC-79106-60: CAPTLOADS Analysis; 24 Aug 83
- (g) GAC Report A51-30-R-79-2; F-14A Aircraft No. 98 Structural Model F-1 Mass Properties Report

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test

F-14A A.C. 98 FATIGUE CONDITIONS

TRANSIENT RESPONSE NON-ARRESTED COND'S (NO THRUST, WT.=51830LBS)

\*\*\*\*\*

FATIGUE COND. NO.	MANEUV.	TIME SLICE	SINK SPEED FT/SEC	DESCRIPTION
1	40F	38	25.3	TAIL DOWN SYMMETRIC
2	40F	76	25.3	TAIL DOWN SYMMETRIC
3	40F	142	25.3	TAIL DOWN SYMMETRIC
4	40F	177	25.3	TAIL DOWN SYMMETRIC
5	32F	38	23.0	TAIL DOWN SYMMETRIC
6	32F	72	23.0	TAIL DOWN SYMMETRIC
7	32F	141	23.0	TAIL DOWN SYMMETRIC
8	32F	185	23.0	TAIL DOWN SYMMETRIC
9	32F	34	18.0	TAIL DOWN SYMMETRIC
10	32F	64	18.0	TAIL DOWN SYMMETRIC
11	32F	119	18.0	TAIL DOWN SYMMETRIC
12	32F	165	18.0	TAIL DOWN SYMMETRIC
13	37F	38	14.0	TAIL DOWN SYMMETRIC
14	37F	68	14.0	TAIL DOWN SYMMETRIC
15	37F	118	14.0	TAIL DOWN SYMMETRIC
16	37F	167	14.0	TAIL DOWN SYMMETRIC
17	35F	80	10.0	TAIL DOWN SYMMETRIC
18	35F	157	10.0	TAIL DOWN SYMMETRIC
19	35F	271	10.0	TAIL DOWN SYMMETRIC
20	35F	41	6.0	TAIL DOWN SYMMETRIC
21	35F	174	6.0	TAIL DOWN SYMMETRIC
22	35F	224	6.0	TAIL DOWN SYMMETRIC
23	71F	32	25.3	LEVEL
24	71F	132	25.3	LEVEL
25	71F	185	25.3	LEVEL
26	71F	211	25.3	LEVEL
27	72F	37	23.0	LEVEL
28	72F	158	23.0	LEVEL
29	72F	215	23.0	LEVEL
30	73F	36	18.0	LEVEL
31	73F	148	18.0	LEVEL
32	73F	213	18.0	LEVEL
33	74F	38	14.0	LEVEL
34	74F	152	14.0	LEVEL
35	74F	202	14.0	LEVEL
36	75F	100	10.0	LEVEL
37	75F	223	10.0	LEVEL
38	75F	108	6.0	LEVEL
39	75F	228	6.0	LEVEL
40	31F	33	28.8	TAIL DOWN ROLLED RIGHT
41	31F	57	28.8	TAIL DOWN ROLLED RIGHT
42	31F	93	28.8	TAIL DOWN ROLLED RIGHT
43	31F	106	28.8	TAIL DOWN ROLLED RIGHT
44	32F	32	23.0	TAIL DOWN ROLLED RIGHT
45	32F	59	23.0	TAIL DOWN ROLLED RIGHT
46	32F	90	23.0	TAIL DOWN ROLLED RIGHT

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test (Continued)

47	82F	115	23.0	TAIL DOWN ROLLED RIGHT
48	83F	75	12.0	TAIL DOWN ROLLED RIGHT
49	83F	52	12.0	TAIL DOWN ROLLED RIGHT
50	83F	85	18.0	TAIL DOWN ROLLED RIGHT
51	83F	110	18.0	TAIL DOWN ROLLED RIGHT
52	84F	77	14.0	TAIL DOWN ROLLED RIGHT
53	84F	63	14.0	TAIL DOWN ROLLED RIGHT
54	84F	110	14.0	TAIL DOWN ROLLED RIGHT
55	84F	120	14.0	TAIL DOWN ROLLED RIGHT
56	85F	51	10.0	TAIL DOWN ROLLED RIGHT
57	85F	106	10.0	TAIL DOWN ROLLED RIGHT
58	86F	73	6.0	TAIL DOWN ROLLED RIGHT
59	86F	152	6.0	TAIL DOWN ROLLED RIGHT

RIGID BODY GROUND COND'S

ARRESTED LANDING (THRUST=2192LBS, WT.=53975LBS, 20DEG SWEEP)

FATIGUE COND. NO.	HOCK DRAG LBS	HOCK SIDE LBS	DESCRIPTION
60	171000	27000	SIDE HOOK FORCE TO LEFT
61	171000	16200	SIDE HOOK FORCE TO LEFT
62	159000	27000	SIDE HOOK FORCE TO LEFT
63	159000	16200	SIDE HOOK FORCE TO LEFT
64	148000	27000	SIDE HOOK FORCE TO LEFT
65	148000	16200	SIDE HOOK FORCE TO LEFT
66	171000	-27000	SIDE HOOK FORCE TO RIGHT
67	171000	-16200	SIDE HOOK FORCE TO RIGHT
68	159000	-27000	SIDE HOOK FORCE TO RIGHT
69	159000	-16200	SIDE HOOK FORCE TO RIGHT
70	148000	-27000	SIDE HOOK FORCE TO RIGHT
71	148000	-16200	SIDE HOOK FORCE TO RIGHT

CATAPULT (WT.=71242LBS, 20DEG SWEEP)

FATIGUE COND. NO.	THRUST LBS	TOW LBS	TORQUE IN-LBS	DESCRIPTION
72	21920	237000		SYMMETRIC TOW-LEVEL
73	21920	213000		SYMMETRIC TOW-LEVEL
74	21920	130000		SYMMETRIC TOW-LEVEL
75	21920	237000	260000	SPOT LEFT
76	21920	213000	260000	SPOT LEFT
77	21920	130000	260000	SPOT LEFT
78	21920	237000	-260000	SPOT RIGHT
79	21920	213000	-260000	SPOT RIGHT
80	21920	130000	-260000	SPOT RIGHT
81		-59100		BUFFERING
82		-59900		BUFFERING
83	21920	-21300		RELEASE

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test (Continued)

UNIT NZ CONDITIONS (BAL. AT FS292 & MAIN GEAR) SWEEP=20

FATIGUE

COND. NO.	DESCRIPTION
84	100( FUEL(16200LBS) ,WT.=63430LBS
85	30( FUEL( 4860LBS) ,WT.=52090LBS

FLIGHT CONDITIONS (WT.=56950 LBS)

FATIGUE

COND. NO.	DESIG.	MACH	ALT (FT)	SWEEP (DEG)	NY	NZ	
86	98-01	.620	10000	25	--	-2.250	SYMM
87	98-02	.620	10000	25	--	3.440	SYMM
88	98-03	.620	10000	25	--	5.346	SYMM
89	98-04	.620	10000	25	--	6.843	SYMM
90	98-05	.620	10000	25	--	7.902	SYMM
91	98-06	.620	10000	25	--	8.243	SYMM
92	98-07	.620	10000	25	--	(1)	ANTI
93	98-08	.620	10000	25	--	(3)	ANTI
94	98-09	.703	10000	35	--	-2.250	SYMM
95	98-10	.703	10000	35	--	3.750	SYMM
96	98-11	.703	10000	35	--	6.500	SYMM
97	98-12	.703	10000	35	--	8.123	SYMM
98	98-13	.703	10000	35	--	9.000	SYMM
99	98-14	.703	10000	35	--	(1)	ANTI
100	98-15	.703	10000	35	--	(3)	ANTI
101	98-16	.785	10000	45	--	-2.250	SYMM
102	98-17	.785	10000	45	--	4.272	SYMM
103	98-18	.785	10000	45	--	9.000	SYMM
104	98-19	.785	10000	45	--	(1)	ANTI
105	98-20	.785	10000	45	--	(3)	ANTI
106	98-21	.868	10000	55	--	-2.250	SYMM
107	98-22	.868	10000	55	--	5.068	SYMM
108	98-23	.868	10000	55	--	8.078	SYMM
109	98-24	.868	10000	55	--	9.000	SYMM
110	98-25	.868	10000	55	--	(1)	ANTI
111	98-26	.868	10000	55	--	(3)	ANTI
112	98-27	.975	10000	68	--	-2.253	SYMM
113	98-28	.975	10000	68	--	6.875	SYMM
114	98-29	.975	10000	68	--	9.000	SYMM
115	98-30	.975	10000	68	--	(1)	ANTI
116	98-31	.975	10000	68	--	(3)	ANTI

VOID 117  
VOID 118  
VOID 119  
VOID 120  
VOID 121  
VOID 122

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test (Continued)

FATIGUE							
COND.							
NO.							
VOID							
124	9E-02	.620	10000	25	--	3.460	SYMM
125	9E-16	.785	10000	45	--	-2.250	SYMM
126	9E-17	.785	10000	45	--	4.272	SYMM
127	9E-27	.975	10000	68	--	-2.253	SYMM
128	9E-28	.975	10000	68	--	6.875	SYMM
129	9E-29	.975	10000	68	--	9.000	SYMM

\*\*\*\*\*

TRANSIENT RESPONSE NON-ARRESTED COND'S (NO THRUST, WT.=51830LBS)

\*\*\*\*\*

FATIGUE	TIME	SINK				
COND.	MANEUV.	SLICE	SPEED	DESCRIPTION		
NO.			FT/SEC			
130	81F	28	25.3	TAIL DOWN	ROLLED	LEFT
131	81F	57	25.3	TAIL DOWN	ROLLED	LEFT
132	81F	80	25.3	TAIL DOWN	ROLLED	LEFT
133	81F	108	25.3	TAIL DOWN	ROLLED	LEFT
134	82F	32	23.0	TAIL DOWN	ROLLED	LEFT
135	82F	59	23.0	TAIL DOWN	ROLLED	LEFT
136	82F	90	23.0	TAIL DOWN	ROLLED	LEFT
137	82F	116	23.0	TAIL DOWN	ROLLED	LEFT
138	83F	35	18.0	TAIL DOWN	ROLLED	LEFT
139	83F	62	18.0	TAIL DOWN	ROLLED	LEFT
140	83F	89	18.0	TAIL DOWN	ROLLED	LEFT
141	83F	110	18.0	TAIL DOWN	ROLLED	LEFT
142	84F	37	14.0	TAIL DOWN	ROLLED	LEFT
143	84F	63	14.0	TAIL DOWN	ROLLED	LEFT
144	84F	110	14.0	TAIL DOWN	ROLLED	LEFT
145	84F	130	14.0	TAIL DOWN	ROLLED	LEFT
146	85F	51	10.0	TAIL DOWN	ROLLED	LEFT
147	85F	106	10.0	TAIL DOWN	ROLLED	LEFT
148	86F	73	6.0	TAIL DOWN	ROLLED	LEFT
149	86F	152	6.0	TAIL DOWN	ROLLED	LEFT

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test (Continued)

CONDITION NUMBERING

SYMMETRIC FLIGHT

REF $\pi_2$	<u>SWEEP ANGLE - DEGREES</u>				
	25	35	45	55	68
-1.75	150*	165*	180*	195*	210*
-1.25	151	166*	181*	196*	211*
-0.75	152	167	182*	197*	212
-0.25	153	168	183	198	213
0.25	154	169	184	199	214
0.75	155	170	185	200	215
1.00	156	171	186	201	216
3.50	157	172	187	202	217
4.50	158	173	188	203	218
5.50	159	174	189	204	219
6.50	160	175	190	205	220
7.50	161	176	191	206	221
8.50	162	177	192	207	222
9.00	163*	178	193	208	223
9.50	164*	179*	194*	209*	224*

\* Values not used

ASYMMETRIC (LEVEL ROLLS)

SWEEP ANGLE PLUS ACCELERATION (1)

$25^\circ + \dot{\rho}$	$35^\circ + \dot{\rho}$	$45^\circ + \dot{\rho}$	$55^\circ + \dot{\rho}$	$68^\circ + \dot{\rho}$
225=155+2(92)	231=171+2(99)	237=186+2(104)	243=201+2(110)	249=216+2(115)
226=156-2(92)	232=171-2(99)	238=186-2(104)	244=201-2(110)	250=216-2(115)
227=156+4(92)	233=171+4(99)	239=186+4(104)	245=201+4(110)	251=216+4(115)
228=156-4(92)	234=171-4(99)	240=186-4(104)	246=201-4(110)	252=216-4(115)
229=156-6(92)	235=171-6(99)	241=186-6(104)	247=201-6(110)	253=216-6(115)
230=156-6(92)	236=171-6(99)	242=186-6(104)	248=201-6(110)	254=216-6(115)

1. Example: New Condition. = 1 g = 9 (unit load)

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
 Fatigue Test (Continued)  
 CONDITION NUMBERING

ASYMMETRIC (TURN REVERSALS)

SWEEP ANGLE PLUS ACCELERATION

$\pi_2$	$25^\circ + \rho$	$35^\circ + \rho$	$45^\circ + \rho$
3.5	255 = 157+1.833(93)	280 = 172+1.833(100)	305 = 187+1.833(105)
	256 = 157-1.833(93)	281 = 172-1.833(100)	306 = 187-1.833(105)
	257 = 157+3.500(93)	282 = 172-3.500(100)	307 = 187+3.500(105)
	258 = 157-3.500(93)	283 = 172-3.500(100)	308 = 187-3.500(105)
	259 = 157-5.167(93)	284 = 172-5.167(100)	309 = 187-5.167(105)
4.5	260 = 157-5.167(93)	285 = 173-5.167(100)	310 = 187-5.167(105)
	261 = 158-1.833(93)	286 = 173+1.833(100)	311 = 188-1.833(105)
	262 = 158-1.833(93)	287 = 173-1.833(100)	312 = 188-1.833(105)
	263 = 158-3.500(93)	288 = 173-3.500(100)	313 = 188-3.500(105)
	264 = 158-3.500(93)	289 = 173-3.500(100)	314 = 188-3.500(105)
	265 = 158-5.167(93)	290 = 173-5.167(100)	315 = 188-5.167(105)
	266 = 158-5.167(93)	291 = 173-5.167(100)	316 = 188-5.167(105)
	267 = 159-1.833(93)	292 = 174-1.833(100)	317 = 189+1.833(105)
	268 = 159-1.833(93)	293 = 174-1.833(100)	318 = 189-1.833(105)
	269 = 159-3.500(93)	294 = 174-3.500(100)	319 = 189-3.500(105)
5.5	270 = 159-3.500(93)	295 = 174-3.500(100)	320 = 189-3.500(105)
	271 = 159-5.167(93)	296 = 174-5.167(100)	321 = 189-5.167(105)
	272 = 159-5.167(93)	297 = 174-5.167(100)	322 = 189-5.167(105)
	273 = 160-2.000(93)	298 = 175-2.000(100)	323 = 190-2.000(105)
	274 = 160-2.000(93)	299 = 175-2.000(100)	324 = 190-2.000(105)
	275 = 160+4.000(93)	300 = 175-4.000(100)	325 = 190-4.000(105)
	276 = 160-4.000(93)	301 = 175-4.000(100)	326 = 190-4.000(105)
6.5	277 = 161-2.000(93)	302 = 176-2.000(100)	327 = 191-2.000(105)
	278 = 161-2.000(93)	303 = 176-2.000(100)	328 = 191-2.000(105)
	279*	304*	329*

\*Values not used

TABLE I Description of Flight and Ground Loading Conditions, Aircraft #98  
Fatigue Test (Continued)

CONDITION NUMBERING

ASYMMETRIC (TURN REVERSALS) (Cont')

SWEEP ANGLE PLUS ACCELERATION

$N_z$	$55^\circ + \dot{p}$	$63^\circ + \dot{p}$
3.5	330 = 202 + 1.833 (111)	355 = 217 + 1.833 (116)
	331 = 202 - 1.833 (111)	356 = 217 - 1.833 (116)
	332 = 202 + 3.500 (111)	357 = 217 + 3.500 (116)
	333 = 202 - 3.500 (111)	358 = 217 - 3.500 (116)
	334 = 202 + 5.167 (111)	359 = 217 + 5.167 (116)
4.5	335 = 202 - 5.167 (111)	360 = 217 - 5.167 (116)
	336 = 203 - 1.833 (111)	361 = 218 - 1.833 (116)
	337 = 203 - 1.833 (111)	362 = 218 - 1.833 (116)
	338 = 203 + 3.500 (111)	363 = 218 + 3.500 (116)
	339 = 203 - 3.500 (111)	364 = 218 - 3.500 (116)
5.5	340 = 203 + 5.167 (111)	365 = 218 + 5.167 (116)
	341 = 203 - 5.167 (111)	366 = 218 - 5.167 (116)
	342 = 204 + 1.833 (111)	367 = 219 - 1.833 (116)
	343 = 204 - 1.833 (111)	368 = 219 - 1.833 (116)
	344 = 204 + 3.500 (111)	369 = 219 + 3.500 (116)
6.5	345 = 204 - 3.500 (111)	370 = 219 - 3.500 (116)
	346 = 204 + 5.167 (111)	371 = 219 + 5.167 (116)
	347 = 204 - 5.167 (111)	372 = 219 - 5.167 (116)
	348 = 205 - 2.000 (111)	373 = 220 + 2.000 (116)
	349 = 205 - 2.000 (111)	374 = 220 - 2.000 (116)
7.5	350 = 205 + 4.000 (111)	375 = 220 + 4.000 (116)
	351 = 205 - 4.000 (111)	376 = 220 - 4.000 (116)
	352 = 206 - 2.000 (111)	377 = 221 - 2.000 (116)
	353 = 206 - 2.000 (111)	378 = 221 - 2.000 (116)
	354*	379 MAXIMUM VALUE 1 - 378
	380 MAXIMUM VALUE 1 - 378	

\* VALUE NOT USED



TABLE 11 Loading Conditions for F 14 Wing Outer Panel Fatigue Test

Condition No.	M/C Test	WOP Test	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	N <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	M <sub>z</sub> (10 <sup>6</sup> )	H <sub>x</sub> (10 <sup>6</sup> )	M <sub>y</sub> (10 <sup>6</sup> )	Description
1	1	1	0	20119	-1.681	0.862	0	1.857	0.349		Landing 25.3 FPS, tail dn symm
2	1	0									
3	1	0									
4	1	0									
5	3	10	0	16972	1.419	0.728	0	-1.567	-0.295		Landing 23.0 FPS, tail dn symm
6	3	0									
7	3	0									
8	3	0									
9	23	76	0	11177	-0.934	0.479	0	-1.032	0.194		Landing 18.0 FPS, tail dn symm
10	23	0									
11	23	0									
12	23	0									
13	51	174	0	7550	-0.632	0.324	0	-0.608	0.131		Landing 14.0 FPS, tail dn symm
14	51	0									
15	51	0									
16	51	0									
17	56	187	0	-4846	-0.405	0.208	0	-0.448	0.084		Landing 10.0 FPS, tail dn symm
18	56	0									
19	58	0									
20	60	216	0	-3037	0.254	0.130	0	-0.280	0.053		Landing 6.0 FPS, tail dn symm
21	66	0									
22	66	0									
23	0	0		same as condition 1							Landing 25.3 FPS, level
24	0	0									
25	0	0									
26	0	0									
27	3	0		same as condition 5							Landing 23.0 FPS, level
28	3	0									
29	3	0									
30	23	0		same as condition 9							Landing 18.0 FPS, level

TABLE II - Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition	Case	POP	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	H <sub>z</sub> (10 <sup>6</sup> )	N <sub>x</sub> (10 <sup>6</sup> )	N <sub>y</sub> (10 <sup>6</sup> )	N <sub>z</sub> (10 <sup>6</sup> )	Description
31	0	0										Landing 14.0 FPS, level
32	0	0										Landing 14.0 FPS, level
33	0	0	Same as condition 13									Landing 10.0 FPS, level
34	0	0										Landing 10.0 FPS, level
35	0	0	Same as condition 17									Landing 6.0 FPS, level
36	0	0										Landing 6.0 FPS, level
37	0	0	Same as condition 20									Landing 25.3 FPS, tail dn roll rt
38	0	0										Landing 25.3 FPS, tail dn roll rt
39	0	0	Same as condition 1									Landing 23.0 FPS, tail dn roll rt
40	0	0										Landing 23.0 FPS, tail dn roll rt
41	0	0	Same as condition 5									Landing 18.0 FPS, tail dn roll rt
42	0	0										Landing 18.0 FPS, tail dn roll rt
43	0	0	Same as condition 9									Landing 14.0 FPS, tail dn roll rt
44	0	0										Landing 14.0 FPS, tail dn roll rt
45	0	0	Same as condition 13									Landing 10.0 FPS, tail dn roll rt
46	0	0										Landing 10.0 FPS, tail dn roll rt
47	0	0	Same as condition 17									Landing 6.0 FPS, tail dn roll rt
48	0	0										Landing 6.0 FPS, tail dn roll rt
49	0	0	Same as condition 20									Arrest, D = 171000, S = 27000, It
50	0	0										Arrest, D = 171000, S = 27000, It

TABLE II - Loading Conditions for F-14 Wing Out-of-Plane Fatigue Test (Continued)

Condition No.	Wing	WOP	$F_x$	$F_y$	$F_z$	$H_x$	$H_y$	$H_z$	$H_x$	$H_y$	$H_z$	Description
						( $10^6$ )	( $10^6$ )	( $10^6$ )	( $10^6$ )	( $10^6$ )	( $10^6$ )	
64	10	0	same as condition 60									Arrest, D = 171000, S = 16200, lt
65	11	112	5950	1/1	0.014	0.007	0.016	0.497	0.016	0.003		Arrest, D = 159000, S = 27000, lt
66	55	0	same as condition 62									Arrest, D = 159000, S = 16200, lt
67	?	0	same as condition 60									
68	10	0	same as condition 60									Arrest, D = 171000, S = 27000, rt
69	?	0	same as condition 60									Arrest, D = 171000, S = 16200, rt
70	10	0	same as condition 60									Arrest, D = 159000, S = 27000, rt
71	10	0	same as condition 60									Arrest, D = 159000, S = 16200, rt
72	32	16	14822	-4597	0.418	0.204	0.459	-1.349	0.459	0.077		Catapult, T = 237000
73	128	64	13432	-4552	-0.414	0.202	-0.455	-1.223	-0.455	0.076		Catapult, T = 213000
74	160	80	11521	-4490	-0.409	0.199	-0.448	-1.049	-0.448	0.075		Catapult, T = 180000
75	32	0										
76	128	0										
77	160	0										
78	32	0										
79	128	0										
80	160	0										
81	160	0										
82	160	0										
83	160	0										
84	320	160	0	-4127	0.376	0.183	0.312	0	0.312	0.069		Static ground, 100% fuel
85	361	361	0	-2019	-0.169	0.087	-0.186	0	-0.186	0.035		Static ground, 30% fuel
86	0	0	0	26418	3.011	1.419	3.322	0	3.322	0.217		25° symm -2.25 R
87	3630	3040	0	43962	5.130	-2.355	5.636	0	5.636	-0.311		25° symm 3.46 R
88	0	0	0	40938	4.575	2.535	5.186	0	5.186	-0.682		25° symm 5.346 R
89	0	0	0	38661	4.075	-2.764	4.804	0	4.804	-1.078		25° symm 6.843 R
90	0	0	0	37442	3.575	3.128	4.473	0	4.473	-1.600		25° symm 7.932 R

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TABLE II Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition	U <sub>z</sub>	U <sub>y</sub>	F <sub>z</sub>	F <sub>y</sub>	(10 <sup>6</sup> ) U <sub>z</sub>	(10 <sup>6</sup> ) U <sub>y</sub>	(10 <sup>6</sup> ) U <sub>z</sub>	(10 <sup>6</sup> ) U <sub>y</sub>	(10 <sup>6</sup> ) U <sub>z</sub>	(10 <sup>6</sup> ) U <sub>y</sub>	Description
94	15	0	3/637	3 499	0	4.306	0	-2.065	25 <sup>o</sup> symm	8.243 g	
95	0	0	9/5	0.059	0	0.118	0	0.016	25 <sup>o</sup> anti	1.0 g	
96	0	0	same as condition 92						25 <sup>o</sup> anti	3.5 g	
97	0	0	22162	1.629	0	2.817	0	0.184	35 <sup>o</sup> symm	-2.25 g	
98	953	0	52359	3.705	0	6.498	0	-0.363	35 <sup>o</sup> symm	3.75 g	
99	42	0	50212	4.018	0	6.356	0	-0.824	35 <sup>o</sup> symm	6.5 g	
100	0	0	50110	-4.441	0	6.232	0	1.390	35 <sup>o</sup> symm	8.123 g	
101	3	0	51362	-4.928	0	6.315	0	1.911	35 <sup>o</sup> symm	9.0 g	
102	0	0	923	0.071	0	-0.111	0	0.016	35 <sup>o</sup> anti	1.0 g	
103	0	0	same as condition 99						35 <sup>o</sup> anti	3.5 g	
104	0	0	23116	1.958	0	-2.868	0	0.077	45 <sup>o</sup> symm	-2.25 g	
105	0	0	56752	-4.831	0	6.974	0	-0.280	45 <sup>o</sup> symm	4.272 g	
106	1	0	63550	-6.136	0	7.510	0	-1.549	45 <sup>o</sup> symm	9.0 g	
107	0	0	-860	0.079	0	-0.102	0	0.014	45 <sup>o</sup> anti	1.0 g	
108	0	0	same as condition 104						45 <sup>o</sup> anti	3.5 g	
109	0	0	-17733	-1.660	0	-2.132	0	-0.014	55 <sup>o</sup> symm	-2.25 g	
110	0	0	58901	-5.529	0	7.158	0	0.117	55 <sup>o</sup> symm	5.068 g	
111	0	0	69723	-7.854	0	9.100	0	-1.177	55 <sup>o</sup> symm	8.078 g	
112	0	0	69781	-8.146	0	9.254	0	-1.453	55 <sup>o</sup> symm	9.0 g	
113	0	0	785	0.080	0	-0.092	0	0.013	55 <sup>o</sup> anti	1.0 g	
114	0	0	same as condition 110						55 <sup>o</sup> anti	3.5 g	
115	0	0	9070	1.139	0	-1.357	0	-0.199	68 <sup>o</sup> symm	-2.253 g	
116	0	0	58374	6.980	0	8.161	0	0.895	68 <sup>o</sup> symm	6.875 g	
117	1	0	67118	-8.048	0	9.248	0	0.694	68 <sup>o</sup> symm	9.0 g	
118	0	0	685	0.074	0	-0.077	0	0.012	68 <sup>o</sup> anti	1.0 g	
119	0	0	same as condition 115						68 <sup>o</sup> anti	3.5 g	
120	0	0									

TABLE 11 Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	Wing	MOP Test	$F_x$	$F_y$	$F_z$	$H_x$ ( $10^6$ )	$H_y$ ( $10^6$ )	$H_z$ ( $10^6$ )	$H_{x'}'$	$H_{y'}'$	Description
131	0	0									
132	0	0									
133	0	0									
134	0	0									
135	0	0									
136	0	0									
137	0	0									
138	0	0									
139	0	0									
130	0	0	same as condition 1								Landing 25.3 FPS, tail dn roll lt
131	0	0									
132	0	0									
133	0	0	same as condition 5								Landing 23.0 FPS, tail dn roll lt
134	0	0									
135	0	0									
136	0	0									
137	0	0									
138	0	0	same as condition 9								Landing 18.0 FPS, tail dn roll lt
139	0	0									
140	0	0									
141	0	0									
142	0	0	same as condition 13								Landing 14.0 FPS, tail dn roll lt
143	0	0									
144	0	0									
145	0	0									
146	0	0	same as condition 17								Landing 10.0 FPS, tail dn roll lt
147	0	0									
148	0	0	same as condition 20								Landing 6.0 FPS, tail dn roll lt
149	0	0									
150	0	0									

TABLE 11 Loading Conditions for F 14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	A/C Test	MOP Test	F <sub>x</sub>	F <sub>z</sub>	M <sub>x</sub> (10 <sup>6</sup> )	M <sub>y</sub> (10 <sup>6</sup> )	M <sub>z</sub> (10 <sup>6</sup> )	N <sub>x</sub> (10 <sup>6</sup> )	N <sub>y</sub> (10 <sup>6</sup> )	N <sub>z</sub> (10 <sup>6</sup> )	Description
151	3	3	0	14092	1.586	0.759	0	-1.754	0.125	0	25° symm -1.25 g
152	9	9	0	2933	-0.869	0.429	0	-0.966	0.081	0	25° symm -0.75 g
153	42	42	0	1769	-0.157	0.099	0	-0.182	0.035	0	25° symm -0.25 g
154	314	314	0	4395	0.055	-0.232	0	0.136	-0.196	0	25° symm 0.25 g
155	2752	2752	0	10559	1.267	-0.562	0	1.385	-0.059	0	25° symm 0.75 g
156	8543	3502	0	13641	1.623	-0.729	0	1.777	-0.083	0	25° symm 1.00 g
157	1722	1722	0	43898	5.118	-2.358	0	5.626	-0.318	0	25° symm 3.50 g
158	812	812	0	42295	4.824	-2.453	0	5.387	-0.514	0	25° symm 4.50 g
159	371	371	0	40703	4.524	-2.559	0	5.147	-0.723	0	25° symm 5.50 g
160	143	143	0	39182	4.190	-2.712	0	4.892	-0.988	0	25° symm 6.50 g
161	57	57	0	37926	3.773	-2.982	0	4.603	-1.392	0	25° symm 7.50 g
162	0	0	0	3798	2.981	-3.806	0	4.168	-2.449	0	25° symm 8.50 g
163	0	0	0								
164	0	0	0								
165	0	0	0								
166	0	0	0								
167	2	2	0	3523	-0.392	0.296	0	-0.489	0.048	0	35° symm -0.75 g
168	13	13	0	2690	0.246	-0.149	0	0.288	0.001	0	35° symm -0.25 g
169	93	93	0	8903	0.884	-0.594	0	1.064	-0.045	0	35° symm 0.25 g
170	816	816	0	15116	1.522	-1.039	0	1.841	-0.091	0	35° symm 0.75 g
171	2100	844	0	18220	1.842	-1.260	0	2.229	-0.112	0	35° symm 1.00 g
172	507	507	0	49282	5.032	-3.483	0	6.110	-0.341	0	35° symm 3.50 g
173	250	240	0	51795	5.250	-3.790	0	6.457	-0.489	0	35° symm 4.50 g
174	110	110	0	51004	5.116	-3.904	0	6.402	-0.656	0	35° symm 5.50 g
175	0	0	0	50212	4.981	-4.018	0	6.346	-0.824	0	35° symm 6.50 g
176	17	17	0	50149	4.739	-4.279	0	6.276	-1.173	0	35° symm 7.50 g
177	4	4	0	50648	4.439	-4.650	0	6.214	-1.646	0	35° symm 8.50 g
178	0	0	0	51362	4.385	-4.928	0	6.314	-1.911	0	35° symm 9.0 g
179	0	0	0								

TABLE II Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	A/C Load	WOP Test	F <sub>z</sub>	F <sub>y</sub>	(10 <sup>6</sup> ) N <sub>x</sub>	(10 <sup>6</sup> ) N <sub>y</sub>	(10 <sup>6</sup> ) N <sub>z</sub>	(10 <sup>6</sup> ) H <sub>x</sub>	(10 <sup>6</sup> ) H <sub>y</sub>	Description
181	0	0	0	0	0.091	-0.124	0	0.150	-0.033	45° synon -0.25 g
182	0	0	0	0	0.638	-0.645	0	0.905	-0.060	45° synon 0.25 g
183	9	9	0	1375	1.185	-1.166	0	1.660	-0.088	45° synon 0.75 g
184	64	64	0	7498	1.458	-1.425	0	2.036	-0.101	45° synon 1.00 g
185	557	557	0	13621	4.193	-4.027	0	5.809	-0.238	45° synon 3.50 g
186	1422	1422	0	16683	5.017	-4.894	0	7.000	-0.341	45° synon 4.50 g
187	345	345	0	47298	4.924	-5.170	0	7.114	-0.610	45° synon 5.50 g
188	165	165	0	57080	4.831	-5.446	0	7.227	-0.878	45° synon 6.50 g
189	75	75	0	58518	4.738	-5.722	0	7.340	-0.146	45° synon 7.50 g
190	29	29	0	59955	4.645	-5.998	0	7.453	-0.415	45° synon 8.50 g
191	12	12	0	61393	4.599	-6.136	0	7.510	-1.549	45° synon 9.0 g
192	3	3	0	62831						
193	0	0	0	63550						
194	0	0	0							
195	0	0	0							
196	0	0	0							
197	0	0	0							
198	7	7	0	3210	0.270	-0.305	0	0.407	0.021	55° synon -0.25 g
199	41	41	0	8447	0.672	-0.796	0	1.041	0.030	55° synon 0.25 g
200	360	360	0	13684	1.074	-1.287	0	1.676	0.039	55° synon 0.75 g
201	936	936	0	16300	1.276	-1.533	0	1.994	0.044	55° synon 1.0 g
202	223	223	0	42480	3.286	-3.989	0	5.167	0.088	55° synon 3.50 g
203	105	105	0	52952	4.091	-4.971	0	6.437	0.107	55° synon 4.50 g
204	48	48	0	60454	4.576	-5.863	0	7.437	-0.069	55° synon 5.50 g
205	18	18	0	64050	4.641	-6.635	0	8.082	-0.499	55° synon 6.50 g
206	7	7	0	67645	4.706	-7.408	0	8.727	-0.929	55° synon 7.50 g
207	2	2	0	69750	4.689	-7.988	0	9.170	-1.304	55° synon 8.50 g
208	0	0	0	69781	4.624	-8.146	0	9.254	-1.453	55° synon 9.0 g
209	0	0	0							
210	0	0	0							

TABLE II Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	3/C 9g	MOP	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	(10 <sup>6</sup> ) U <sub>x</sub>	(10 <sup>6</sup> ) U <sub>y</sub>	(10 <sup>6</sup> ) N <sub>z</sub>	(10 <sup>6</sup> ) U <sub>z</sub>	(10 <sup>6</sup> ) H <sub>y</sub>	Description
211	0	0	2068	-0.198	0	0.073	-0.198	0	0.210	-0.019	68° symm
212	4	4	5757	-0.643	0	0.353	-0.643	0	0.732	0.041	68° symm
213	9	9	9446	-1.088	0	0.631	-1.088	0	1.254	0.101	68° symm
214	69	69	13135	-1.533	0	0.909	-1.533	0	1.775	0.160	68° symm
215	600	600	14980	-1.754	0	1.049	-1.754	0	2.035	0.192	68° symm
216	1518	613	33424	-3.978	0	2.441	-3.978	0	4.641	0.490	68° symm
217	371	371	40802	-4.868	0	2.999	-4.868	0	5.685	0.611	68° symm
218	181	181	48180	-5.757	0	3.556	-5.757	0	6.727	0.731	68° symm
219	83	83	55557	-6.646	0	4.113	-6.646	0	7.769	0.851	68° symm
220	31	31	60910	-7.294	0	4.406	-7.294	0	8.480	0.836	68° symm
221	12	12	65049	-7.794	0	4.541	-7.794	0	8.990	0.743	68° symm
222	3	3	67118	-8.048	0	4.608	-8.048	0	9.248	0.694	68° symm
223	0	0									
224	0	0	15591	-0.611	0	1.415	-0.611	0	1.540	-0.050	25° level roll 2 rad/sec 2
225	0	0	11691	-0.847	0	1.831	-0.847	0	2.014	-0.117	25° level roll -2 rad/sec 2
226	0	0	17541	-0.493	0	1.207	-0.493	0	1.304	-0.016	25° level roll 4 rad/sec 2
227	376	376	9741	-0.965	0	2.039	-0.965	0	2.251	-0.151	25° level roll -4 rad/sec 2
228	376	376	19491	-0.375	0	0.999	-0.375	0	1.067	0.017	25° level roll 6 rad/sec 2
229	143	143	7791	-1.083	0	2.247	-1.083	0	2.488	-0.184	25° level roll -6 rad/sec 2
230	143	143	20066	-1.118	0	1.668	-1.118	0	2.006	-0.082	35° level roll 2 rad/sec 2
231	0	0	16374	-1.402	0	2.016	-1.402	0	2.451	-0.142	35° level roll -2 rad/sec 2
232	0	0	21917	-0.976	0	1.494	-0.976	0	1.784	-0.052	35° level roll 4 rad/sec 2
233	111	111	14528	-1.544	0	2.190	-1.544	0	2.674	-0.172	35° level roll -4 rad/sec 2
234	111	111	23758	-0.834	0	1.320	-0.834	0	1.501	-0.021	35° level roll 6 rad/sec 2
235	42	42	12687	-1.686	0	2.364	-1.686	0	2.897	-0.202	35° level roll -6 rad/sec 2
236	42	42	18403	-1.267	0	1.324	-1.267	0	1.831	-0.072	45° level roll 2 rad/sec 2
237	0	0	14963	-1.583	0	1.592	-1.583	0	2.241	-0.131	45° level roll -2 rad/sec 2
238	0	0	20123	-1.109	0	1.190	-1.109	0	1.626	-0.042	45° level roll 4 rad/sec 2
239	//	//	13243	-1.741	0	1.726	-1.741	0	2.446	-0.160	45° level roll -4 rad/sec 2
240	//	//									



TABLE II Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition	$V_{\text{CR}}^2$ rad/sec	WOP Test	$F_x$	$F_z$	$H_x$ ( $10^6$ )	$H_y$ ( $10^6$ )	$H_z$ ( $10^6$ )	$H_x$ ( $10^6$ )	$H_y$ ( $10^6$ )	Description
241	29	29	0	21863	1.056	0.951	0	1.471	0.013	45° level roll 6 rad/sec <sup>2</sup>
242	29	29	0	11523	1.860	-1.899	0	2.651	-0.190	45° level roll -6 rad/sec <sup>2</sup>
243	0	0	0	17876	1.182	-1.373	0	1.810	0.070	55° level roll 2 rad/sec <sup>2</sup>
244	0	0	0	14724	1.370	-1.693	0	2.178	0.018	55° level roll -2 rad/sec <sup>2</sup>
245	48	48	0	19452	1.088	-1.213	0	1.627	0.096	55° level roll 4 rad/sec <sup>2</sup>
246	48	48	0	13148	1.464	-1.853	0	2.362	-0.008	55° level roll -4 rad/sec <sup>2</sup>
247	18	18	0	21028	0.994	-1.053	0	1.443	0.122	55° level roll 6 rad/sec <sup>2</sup>
248	18	18	0	11572	1.558	-2.013	0	2.545	-0.034	55° level roll -6 rad/sec <sup>2</sup>
249	0	0	0	16350	1.003	-1.606	0	1.881	0.214	68° level roll 2 rad/sec <sup>2</sup>
250	0	0	0	13610	1.095	-1.902	0	2.188	0.169	68° level roll -2 rad/sec <sup>2</sup>
251	84	84	0	17720	0.957	-1.458	0	1.728	0.236	68° level roll 4 rad/sec <sup>2</sup>
252	84	84	0	12240	1.141	-2.05	0	2.342	0.147	68° level roll -4 rad/sec <sup>2</sup>
253	32	32	0	19090	0.911	-1.310	0	1.575	0.258	68° level roll 6 rad/sec <sup>2</sup>
254	32	32	0	10870	1.187	-2.198	0	2.495	-0.125	68° level roll -6 rad/sec <sup>2</sup>
255	464	464	0	45685	4.927	-2.250	0	5.409	-0.288	25° turn, 3.5 g & 1.833
256	464	464	0	42111	5.309	-2.466	0	5.843	-0.349	25° turn, 3.5 g & -1.833
257	47	47	0	47311	4.754	-2.152	0	5.212	-0.260	25° turn, 3.5 g & 3.500
258	47	47	0	40486	5.482	-2.565	0	6.040	-0.377	25° turn, 3.5 g & -3.500
259	11	11	0	48936	4.581	-2.053	0	5.014	-0.232	25° turn, 3.5 g & 5.167
260	11	11	0	38860	5.655	-2.663	0	6.238	-0.405	25° turn, 3.5 g & -5.167
261	317	317	0	44082	4.633	-2.345	0	5.170	-0.484	25° turn, 4.5 g & 1.833
262	317	317	0	40508	5.015	-2.561	0	5.604	-0.545	25° turn, 4.5 g & -1.833
263	37	37	0	45708	4.460	-2.247	0	4.973	-0.456	25° turn, 4.5 g & 3.500
264	37	37	0	38883	5.188	-2.660	0	5.802	-0.573	25° turn, 4.5 g & -3.500
265	9	9	0	47333	4.287	-2.148	0	4.776	-0.428	25° turn, 4.5 g & 5.167
266	9	9	0	37257	5.361	-2.758	0	5.999	-0.601	25° turn, 4.5 g & -5.167
267	123	123	0	42490	4.333	-2.451	0	4.930	-0.692	25° turn, 5.5 g & 1.833
268	123	123	0	38916	4.715	-2.667	0	5.364	-0.754	25° turn, 5.5 g & -1.833
269	16	16	0	44116	4.160	-2.353	0	4.733	-0.664	25° turn, 5.5 g & 3.500
270	16	16	0	37291	4.888	-2.766	0	5.561	-0.782	25° turn, 5.5 g & -3.500

TABLE II - Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	Case	Y <sub>z</sub>	Wing Test	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	H <sub>z</sub> (10 <sup>6</sup> )	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	H <sub>z</sub> (10 <sup>6</sup> )	Description
271	4	4	4	0	45741	3.987	2.254	0	4.535	-0.636	25 <sup>0</sup> turn, 5.5 g & 5.167		
272	4	4	4	0	35665	5.061	-2.864	0	5.759	-0.810	25 <sup>0</sup> turn, 5.5 g & -5.167		
273	36	36	36	0	41132	3.982	-2.594	0	4.656	-0.954	25 <sup>0</sup> turn, 6.5 g & 2.000		
274	36	36	36	0	37232	4.398	-2.830	0	5.129	-1.021	25 <sup>0</sup> turn, 6.5 g & -2.000		
275	6	6	6	0	43082	3.774	-2.476	0	4.419	-0.921	25 <sup>0</sup> turn, 6.5 g & 4.000		
276	6	6	6	0	35282	4.606	-2.948	0	5.366	-1.055	25 <sup>0</sup> turn, 6.5 g & -4.000		
277	10	10	10	0	39876	3.565	2.864	0	4.367	-1.358	25 <sup>0</sup> turn, 7.5 g & 2.000		
278	10	10	10	0	35976	3.981	-3.100	0	4.650	-1.425	25 <sup>0</sup> turn, 7.5 g & -2.000		
279	0	0	0	0									
280	17	17	17	0	50974	4.873	3.353	0	5.906	0.313	35 <sup>0</sup> turn, 3.5 g & 1.833		
281	17	17	17	0	47590	5.191	-3.613	0	6.314	-0.368	35 <sup>0</sup> turn, 3.5 g & -1.833		
282	14	14	14	0	52513	4.728	-3.235	0	5.721	-0.288	35 <sup>0</sup> turn, 3.5 g & 3.500		
283	14	14	14	0	46052	5.337	-3.732	0	6.500	-0.394	35 <sup>0</sup> turn, 3.5 g & -3.500		
284	3	3	3	0	54051	4.582	-3.116	0	5.535	-0.263	35 <sup>0</sup> turn, 3.5 g & 5.167		
285	3	3	3	0	44513	5.482	-3.850	0	6.685	-0.419	35 <sup>0</sup> turn, 3.5 g & -5.167		
286	94	94	94	0	53487	5.091	-3.660	0	6.253	-0.461	35 <sup>0</sup> turn, 4.5 g & 1.833		
287	94	94	94	0	50103	5.409	-3.920	0	6.661	-0.516	35 <sup>0</sup> turn, 4.5 g & -1.833		
288	11	11	11	0	55026	4.946	-3.542	0	6.067	-0.436	35 <sup>0</sup> turn, 4.5 g & 3.500		
289	11	11	11	0	48565	5.555	-4.039	0	6.846	-0.541	35 <sup>0</sup> turn, 4.5 g & -3.500		
290	2	2	2	0	56564	4.800	-3.423	0	5.882	-0.411	35 <sup>0</sup> turn, 4.5 g & 5.167		
291	2	2	2	0	47026	5.700	4.157	0	7.032	-0.567	35 <sup>0</sup> turn, 4.5 g & -5.167		
292	36	36	36	0	52696	4.957	-3.774	0	6.198	-0.628	35 <sup>0</sup> turn, 5.5 g & 1.833		
293	36	36	36	0	49312	5.275	-4.034	0	6.606	-0.683	35 <sup>0</sup> turn, 5.5 g & -1.833		
294	5	5	5	0	54235	4.812	-3.656	0	6.012	-0.603	35 <sup>0</sup> turn, 5.5 g & 3.500		
295	5	5	5	0	47774	5.421	-4.153	0	6.791	-0.709	35 <sup>0</sup> turn, 5.5 g & -3.500		
296	1	1	1	0	55773	4.666	-3.537	0	5.827	-0.578	35 <sup>0</sup> turn, 5.5 g & 5.167		
297	1	1	1	0	46235	5.566	-4.271	0	6.977	-0.734	35 <sup>0</sup> turn, 5.5 g & -5.167		
298	10	10	10	0	52058	4.807	-3.876	0	6.124	-0.793	35 <sup>0</sup> turn, 6.5 g & 2.000		
299	10	10	10	0	48366	5.155	-4.160	0	6.569	-0.854	35 <sup>0</sup> turn, 6.5 g & -2.000		
300	2	2	2	0	53904	4.633	-3.734	0	5.901	-0.763	35 <sup>0</sup> turn, 6.5 g & 4.000		

TABLE II - Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	V/C Test	WOP Test	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	(10 <sup>6</sup> ) M <sub>x</sub>	(10 <sup>6</sup> ) M <sub>y</sub>	(10 <sup>6</sup> ) M <sub>z</sub>	(10 <sup>6</sup> ) M <sub>xy</sub>	(10 <sup>6</sup> ) M <sub>yz</sub>	Description
301	2	0	46520	5,329	-4,302	0	6,791	-0.884	35° turn, 6.5 g & 4.000		
302	3	0	51995	4,565	-4,137	0	6,054	-1.142	35° turn, 7.5 g & 2.000		
303	3	0	48303	4,913	-4,421	0	6,499	-1.203	35° turn, 7.5 g & -2.000		
304	0	0									
305	91	91	48874	4,070	3,882	0	5,621	-0.211	45° turn, 3.5 g & 1.833		
306	91	91	45722	4,316	-4,172	0	5,997	-0.265	45° turn, 3.5 g & -1.833		
307	9	9	50308	3,959	-3,751	0	5,450	-0.186	45° turn, 3.5 g & 3.500		
308	9	9	44288	4,428	-4,304	0	6,168	-0.290	45° turn, 3.5 g & -3.500		
309	2	2	51742	3,847	-3,619	0	5,279	-0.162	45° turn, 3.5 g & 5.167		
310	2	2	42854	4,539	-4,435	0	6,338	-0.314	45° turn, 3.5 g & -5.167		
311	64	64	58656	4,894	-4,749	0	6,812	-0.314	45° turn, 4.5 g & 1.833		
312	64	64	55504	5,140	-5,039	0	7,188	-0.368	45° turn, 4.5 g & -1.833		
313	8	8	60090	4,783	-4,618	0	6,642	-0.290	45° turn, 4.5 g & 3.500		
314	8	8	54070	5,252	-5,171	0	7,359	-0.393	45° turn, 4.5 g & -3.500		
315	2	2	61524	4,671	-4,486	0	6,471	-0.265	45° turn, 4.5 g & 5.167		
316	2	2	52636	5,363	-5,302	0	7,530	-0.418	45° turn, 4.5 g & -5.16		
317	25	25	60094	4,801	-5,025	0	6,926	-0.583	45° turn, 5.5 g & 1.833		
318	25	25	56942	5,047	-5,315	0	7,302	-0.637	45° turn, 5.5 g & -1.833		
319	3	3	61528	4,690	-4,894	0	6,755	-0.558	45° turn, 5.5 g & 3.500		
320	3	3	55508	5,159	-5,447	0	7,472	-0.661	45° turn, 5.5 g & -3.500		
321	1	1	62967	4,578	-4,762	0	6,584	-0.533	45° turn, 5.5 g & 5.167		
322	1	1	54074	5,270	-5,578	0	7,643	-0.686	45° turn, 5.5 g & -5.167		
323	8	8	61675	4,697	-5,288	0	7,022	-0.849	45° turn, 6.5 g & 2.000		
324	8	8	58235	4,965	-5,604	0	7,432	-0.908	45° turn, 6.5 g & -2.000		
325	1	1	63395	4,563	-5,130	0	6,817	-0.819	45° turn, 6.5 g & 4.000		
326	1	1	56515	5,099	-5,762	0	7,637	-0.937	45° turn, 6.5 g & -4.000		
327	2	2	63113	4,604	-5,564	0	7,135	-1.117	45° turn, 7.5 g & 2.000		
328	2	2	59673	4,872	-5,880	0	7,545	-1.176	45° turn, 7.5 g & -2.000		
329	0	0									
330	60	60	43924	3,200	-3,842	0	4,999	-0.112	55° turn, 3.5 g & 1.833		

TABLE II Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Continued)

Condition No.	Y/C RB Test	MOP Test	F <sub>x</sub>	F <sub>y</sub>	F <sub>z</sub>	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	M <sub>z</sub> (10 <sup>6</sup> )	H <sub>x</sub> ' (10 <sup>6</sup> )	H <sub>y</sub> ' (10 <sup>6</sup> )	Description
331	0	60	0	41036	0	3.372	-4.136	0	5.336	0.064	55° turn, 3.5 g & -1.833
332	0	6	0	45238	0	3.122	-3.709	0	4.856	0.134	55° turn, 3.5 g & 3.500
333	0	6	0	39222	0	3.451	-4.269	0	5.389	0.043	55° turn, 3.5 g & 3.500
335	1	1	0	46552	0	3.053	3.576	0	4.693	0.155	55° turn, 3.5 g & 5.167
335	1	1	0	38408	0	3.529	-4.402	0	5.642	0.021	55° turn, 3.5 g & -5.167
336	31	31	0	54396	0	4.005	-4.824	0	6.269	0.131	55° turn, 4.5 g & 1.833
337	31	31	0	51508	0	4.177	-5.118	0	6.605	0.083	55° turn, 4.5 g & -1.833
338	5	5	0	55710	0	3.927	-4.691	0	6.116	0.152	55° turn, 4.5 g & 3.500
339	5	5	0	50194	0	4.256	-5.251	0	6.759	0.061	55° turn, 4.5 g & -3.500
340	1	1	0	57024	0	3.848	-4.558	0	5.962	0.174	55° turn, 4.5 g & 5.167
341	1	1	0	48880	0	4.334	-5.384	0	6.912	0.039	55° turn, 4.5 g & -5.167
342	16	16	0	61898	0	4.490	-5.716	0	7.269	-0.045	55° turn, 5.5 g & 1.833
343	16	16	0	59010	0	4.662	-6.010	0	7.605	-0.093	55° turn, 5.5 g & -1.833
345	2	2	0	63212	0	4.412	-5.583	0	7.116	-0.023	55° turn, 5.5 g & 3.500
345	2	2	0	57696	0	4.741	-6.143	0	7.759	-0.115	55° turn, 5.5 g & -3.500
346	0	0	0	64526	0	4.333	-5.450	0	6.962	-0.002	55° turn, 5.5 g & 5.167
347	0	0	0	56382	0	4.819	-6.276	0	7.912	-0.136	55° turn, 5.5 g & -5.167
348	4	4	0	65626	0	4.547	-6.475	0	7.898	-0.473	55° turn, 6.5 g & 2.000
349	4	4	0	62474	0	4.735	-6.795	0	8.265	-0.525	55° turn, 6.5 g & -2.000
350	1	1	0	67202	0	4.553	-6.315	0	7.714	-0.447	55° turn, 6.5 g & 4.000
351	1	1	0	60898	0	4.829	-6.955	0	8.449	-0.551	55° turn, 6.5 g & -4.000
352	1	1	0	69221	0	4.612	-7.248	0	8.543	-0.903	55° turn, 7.5 g & 2.000
353	1	1	0	66069	0	4.800	-7.568	0	8.911	-0.955	55° turn, 7.5 g & -2.000
354	0	0	0		0			0			
355	95	95	0	34680	0	2.399	-3.842	0	4.501	0.511	68° turn, 3.5 g & 1.833
356	95	95	0	32168	0	2.483	-4.114	0	4.782	0.470	68° turn, 2.5 g & -1.833
357	10	10	0	35822	0	2.361	3.719	0	4.373	0.529	68° turn, 3.5 g & 3.500
358	10	10	0	31027	0	2.522	-4.237	0	4.910	0.451	68° turn, 3.5 g & -3.500
359	2	2	0	36963	0	2.322	-3.596	0	4.255	0.548	68° turn, 3.5 g & 5.167
360	2	2	0	29885	0	2.560	-4.360	0	5.036	0.433	68° turn, 3.5 g & -5.167

TABLE II - Loading Conditions for F-14 Wing Outer Panel Fatigue Test (Cont. Inued)

Condition	N/C %	WOP	F	F <sub>z</sub>	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	H <sub>z</sub> (10 <sup>6</sup> )	H <sub>x</sub> (10 <sup>6</sup> )	H <sub>y</sub> (10 <sup>6</sup> )	Description
36.1	71	71	0	42057	2.957	4.732	0	5.544	0.631	68° turn, 4.5 g & 1.833
36.2	71	71	0	39546	3.041	5.004	0	5.876	0.590	68° turn, 4.5 g & -1.833
36.3	8	8	0	43200	2.919	-4.609	0	5.417	0.650	68° turn, 4.5 g & 3.500
36.4	8	8	0	38405	3.030	-5.127	0	5.953	0.572	68° turn, 4.5 g & -3.500
36.5	2	2	0	44341	2.880	4.486	0	5.789	0.668	68° turn, 4.5 g & 5.167
36.6	2	2	0	37263	3.118	-5.250	0	6.081	0.553	68° turn, 4.5 g & -5.167
36.7	28	28	0	49436	3.514	-5.621	0	6.587	0.751	68° turn, 5.5 g & 1.833
36.8	28	28	0	46924	3.598	-5.893	0	6.808	0.710	68° turn, 5.5 g & -1.833
36.9	3	3	0	50578	3.476	5.498	0	6.459	0.769	68° turn, 5.5 g & 3.500
37.0	3	3	0	45783	3.637	-6.016	0	6.996	0.692	68° turn, 5.5 g & -3.500
37.1	1	1	0	51719	3.437	-5.375	0	6.331	0.788	68° turn, 5.5 g & 5.167
37.2	1	1	0	44641	3.675	-6.139	0	7.123	0.673	68° turn, 5.5 g & -5.167
37.3	8	8	0	56927	4.067	-6.498	0	7.616	0.873	68° turn, 6.5 g & 2.000
37.4	8	8	0	54187	4.159	-6.794	0	7.973	0.828	68° turn, 6.5 g & -2.000
37.5	1	1	0	58297	4.021	-6.350	0	7.503	0.895	68° turn, 6.5 g & 4.000
37.6	1	1	0	52817	4.205	-6.942	0	8.076	0.806	68° turn, 6.5 g & -4.000
37.7	2	2	0	62280	4.360	-7.146	0	8.327	0.858	68° turn, 7.5 g & 2.000
37.8	2	2	0	59540	4.452	-7.442	0	8.634	0.814	68° turn, 7.5 g & -2.000

Table III Calculation of Net Wing Pivot Loads for Symmetric Flight Conditions

Condition	$\delta$	$\theta$	$\alpha$	$\beta$	$\gamma$	$F_z/g$	$M_x/g$	$M_y/g$	$F_z_{AIR}$	$M_{x_{AIR}}$	$M_{y_{AIR}}$	$F_z$	$M_{x_{AIR}}$	$M_{y_{AIR}}$	$F_z$	$M_x \times 10^6$	$M_y \times 10^6$	$M_x \times 10^6$	$M_y \times 10^6$
87	3.00	7.15	391.91	79	8600	4200	30961	-3.370	-1.666	0.359	-0.227	-26418	-3.011	1.419	-3.321	0.217			
88	3.00	6.20	391.91	130	10500	6900	50948	5.683	-2.704	-0.553	0.349	43962	5.130	-2.355	5.636	-0.311			
89	3.00	16.0	391.91	132	10522	7844	51732	5.429	-3.074	0.854	0.539	40938	4.575	-2.535	5.186	-0.682			
90	6.00	25.0	391.91	133	13187	-8812	52477	5.168	-3.434	-1.093	0.690	38661	4.075	-2.764	4.804	-1.078			
91	7.00	35.0	391.91	136	12356	10022	53457	4.852	-3.928	1.267	0.800	37442	3.575	-3.128	4.473	-1.60			
92	8.00	42.5	391.91	136.5	11050	11050	54280	4.506	-4.331	1.316	0.832	37637	3.250	-3.499	4.306	-2.064			
93	9.00	50.3	503.864	53	5200	3000	-26705	-2.620	1.915	0.314	-0.286	-22162	-2.306	1.629	-2.820	0.184			
94	9.00	7.6	503.864	119	11700	8100	59960	5.895	-6.182	-0.524	0.477	52389	5.351	-3.705	6.50	-0.363			
95	9.00	16.0	503.864	125.7	11250	9613	63336	5.807	-4.864	-0.969	0.826	50212	4.981	-4.018	6.350	-0.824			
96	8.00	25.0	503.864	132	11158	10863	66510	5.723	-5.473	-1.135	1.032	50110	4.588	-4.441	6.232	-1.390			
97	9.00	31.6	503.864	138	11500	-12050	69533	5.643	-6.072	-1.258	1.144	51362	4.385	-4.928	6.314	-1.911			
101	9.00	7.7	628.624	44	3750	3050	-27659	-2.337	2.294	0.260	-0.376	-23116	-2.097	1.958	-2.868	0.077			
102	9.00	7.8	628.624	104	8500	8700	65377	5.532	-5.469	-0.494	0.638	56752	5.038	-4.831	6.974	-0.280			
103	9.00	17.5	628.624	130	8970	-11900	81721	5.639	7.481	-0.040	1.345	63550	4.599	-6.136	7.510	-1.549			
106	9.00	2.2	768.143	29	2000	2650	-22276	-1.536	2.036	0.198	-0.376	-17733	-1.338	1.660	-2.132	-0.014			
107	9.00	8.2	768.143	90	6500	-8300	69133	4.993	-6.376	-0.445	0.847	58901	4.548	-5.529	7.158	0.117			
108	9.00	14.0	768.143	112	7100	-10400	86032	5.434	-7.989	-0.710	0.135	69723	4.744	-7.854	9.10	-1.177			
109	9.00	15.75	768.143	114.5	7050	-10800	87952	5.415	-8.296	-0.791	0.150	69781	4.624	-8.146	9.253	-1.453			
112	9.00	2.25	969.196	14	-900	1600	-13569	-0.872	1.551	0.168	-0.412	-9020	-0.764	1.139	-1.357	-0.199			
113	9.00	3.5	969.196	74.5	4600	-8500	72205	4.652	-8.218	-1.363	0.330	58324	4.322	-6.980	8.161	0.895			
114	9.00	13.1	969.196	88	5200	-10000	85289	5.040	-9.692	-0.432	1.644	67118	4.608	-8.048	9.248	0.694			

TABLE IV VARIATION OF WING CENTER OF PRESSURE LOCATION WITH SWEEP ANGLE

Wing Sweep Angle, Degrees	X <sub>CP</sub> , Inches	Y <sub>CP</sub> , Inches
15°	122.5	2.1
35°	121.0	2.1
45°	120.0	0.0
55°	115.9	-4.0
65°	134.0	-16.1

TABLE V SUMMARY OF C.G. LOAD FACTORS FOR ARRESTED LANDING CONDITIONS:  
AIRCRAFT NO. 98 FATIGUE TEST

Condition No.	Description	Drag, lb	$N_x$	$N_z$
60, 61, 66, 67				
64, 65, 70, 71	Upper 90%	171000	3.169	-.0913
62, 63, 68, 69	Mean	159000	2.947	-.0849

Notes: - Taken from reference (c)  
 -  $N_x$  and angular accelerations omitted  
 - 173,000 lb conditions (64, 65, 70, 71) combined  
 with 171,000 lb conditions per NAVAIR direction



TABLE VI SUMMARY OF C.G. LOAD FACTORS FOR CATAPULT CONDITIONS:  
AIRCRAFT NO. 98 FATIGUE TEST

Condition No.	Description	$M_x$	$M_y$
72	Tow = 237000 lb	3.592	1.114
73	Tow = 213000 lb	3.055	1.103
74	Tow = 180000 lb	2.792	1.088

Notes: Taken from reference (c)  
-  $M_y$  and angular accelerations zero

TABLE VII - SUMMARY OF EVENTS INCLUDED IN F-14 WING OUTER PANEL FATIGUE TEST SPECTRUM  
 BASED ON A BLOCK SIZE OF 440 FLIGHTS (480 HOURS)

FLIGHT

Sweep	<u>MANEUVERS VERSUS WING SWEEP</u>					
	1	2	3	4	5	6
	Symmetric	Turn Reversals	Level Rolls	Asymmetric 2 + 3	Maneuvers at N > 1.0 1 <sup>2</sup> + 2	Total Maneuvers 1 + 4
25	2040	1080	519	1599	3120	3639
35	616	308	153	461	924	1077
45	414	216	106	322	630	736
55	270	138	66	204	408	474
68	521	231	116	347	682	798
Sum	3791	1973	960	2933	5764	6724

LANDINGS

STRENGTH VERSUS CYCLES

Sink Speed	Cycles
25.0	1
25.0	10
18.0	76
14.0	174
10.0	187
6.0	216
Sum	664

CATAPULTS

LAUNCH (TOW) LOAD VERSUS CYCLES

Load	Cycles
237000	16
213000	64
180000	80
Sum	160

ARRESTMENTS

ARRESTMENT LOAD (DRAG) VERSUS CYCLES

Load	Cycles
171000	48
159000	112
Sum	160

TABLE VIII - Flight Data For Wing Sweep Change Under Load

CHANGE IN WING SWEEP AT VARIOUS  
LEVELS OF WING BEARING AXIAL LD  
CYCLES FOR 185.2 HOURS

BEARING LOAD RANGE < LD ≤	WING SWEEP CHANGE											
	0.3	2.0	5.0	10.0	15.0	20.0	25.0	30.0	35.0	40.0	45.0	50.0
0.	57	26	8	2	~	~	~	~	1	~	~	98
50000.	500	112	92	10	5	1	4	2	~	1	1	~
100000.	3261	703	411	105	56	35	13	19	22	16	16	~
150000.	2869	401	169	46	19	14	7	1	~	1	3	~
200000.	1065	195	69	23	9	9	2	3	4	~	~	~
250000.	765	160	53	11	3	5	~	1	~	~	~	~
300000.	741	156	39	7	1	~	~	~	~	~	~	~
350000.	652	191	50	5	2	~	2	~	~	~	~	~
400000.	603	193	54	10	2	1	~	~	~	~	~	~
450000.	455	182	36	1	~	~	~	~	~	~	~	~
500000.	298	125	25	1	1	~	~	~	~	~	~	~
550000.	92	27	3	~	~	~	~	~	~	~	~	~
600000.	20	9	1	~	~	~	~	~	~	~	~	~
650000.	2	1	~	~	~	~	~	~	~	~	~	~
700000.	~	~	~	~	~	~	~	~	~	~	~	~
750000.	~	~	~	~	~	~	~	~	~	~	~	~
800000.	~	~	~	~	~	~	~	~	~	~	~	~
850000.	~	~	~	~	~	~	~	~	~	~	~	~
900000.	~	~	~	~	~	~	~	~	~	~	~	~

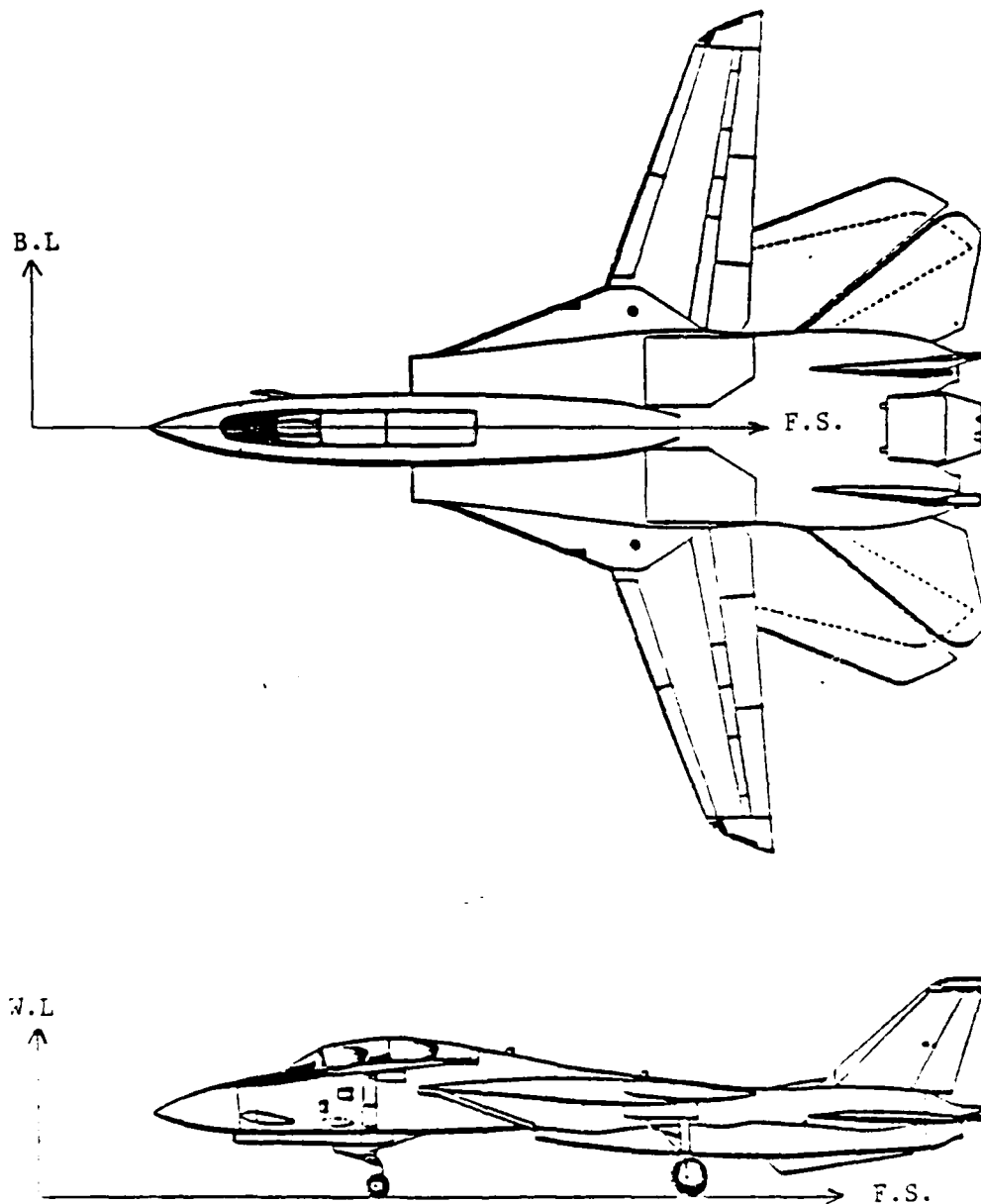
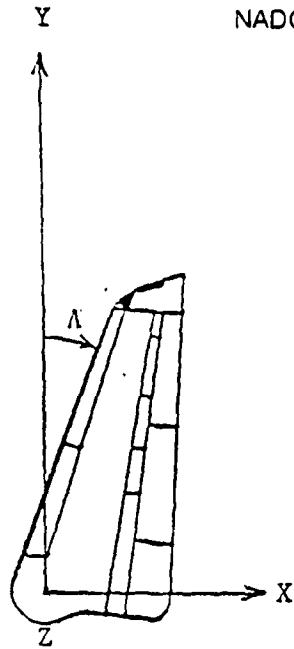


Figure 1. Axes and Sign Conventions

a. Aircraft Axis System



Origin at Wing Pivot Axis

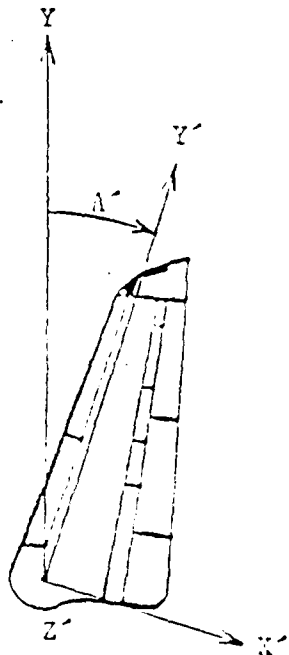
F.S. = 524.203

B.L. = 107.000

W.L. = 145.000

Forces positive as shown  
Right hand rule for moments

b. Unswept Wing Axis System



$$A' = A - 3.5^\circ$$

c. Wing Outer Panel Axis System

Figure 1. Axes and Sign Conventions Cont.

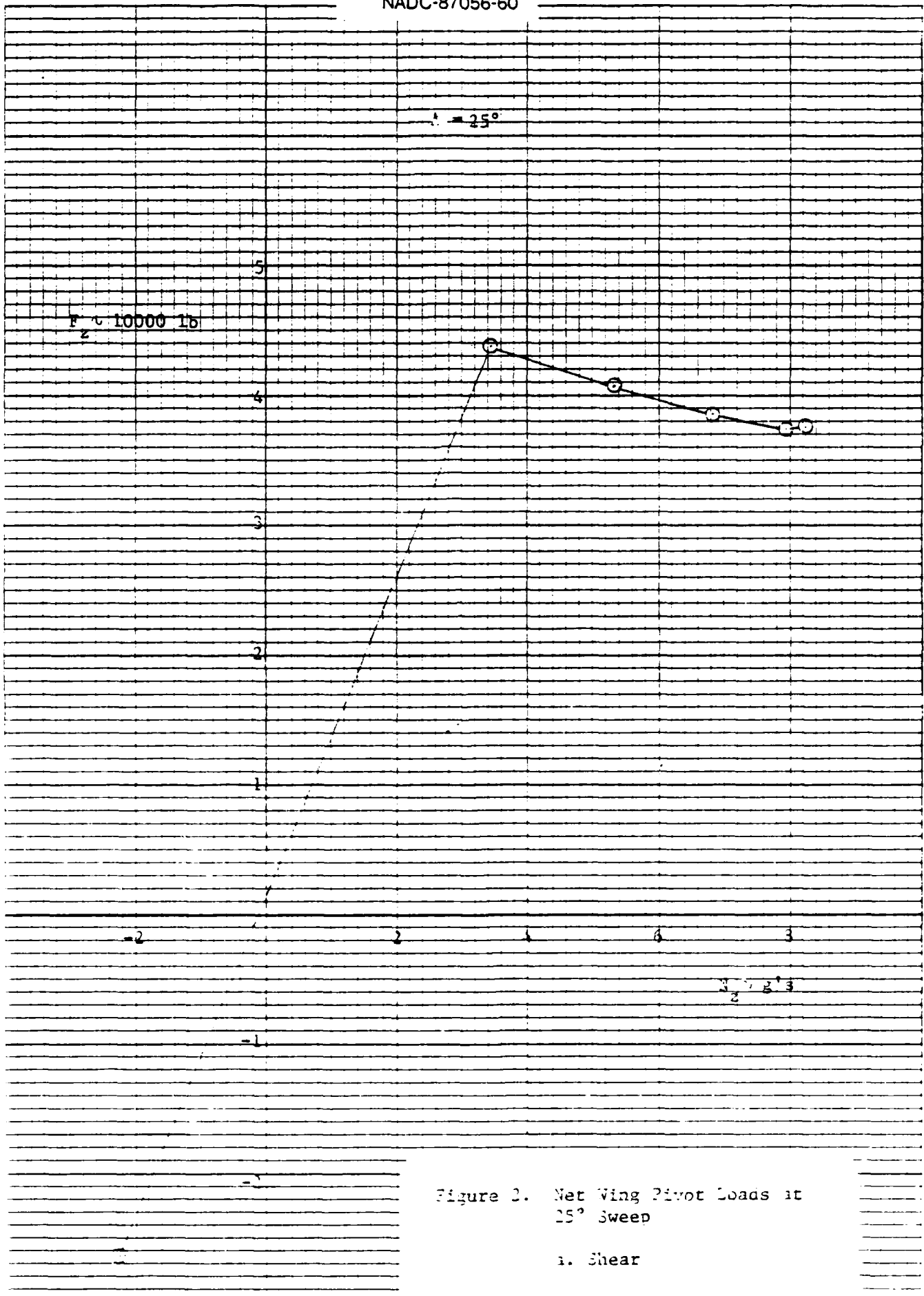


Figure 2. Net Wing Pivot Loads at 25° Sweep

a. Shear

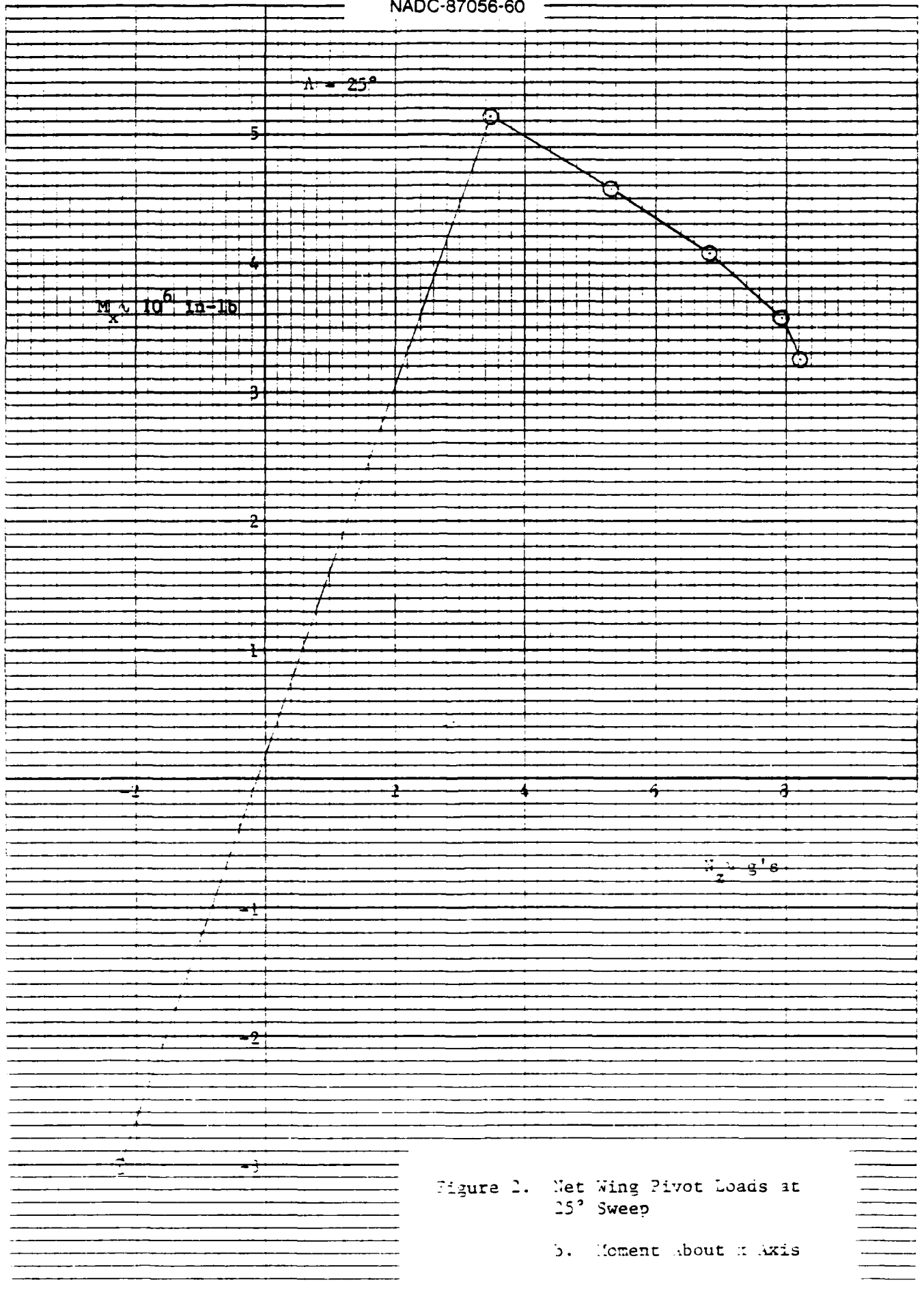


Figure 2. Net Wing Pivot Loads at 25° Sweep

b. Moment about x Axis

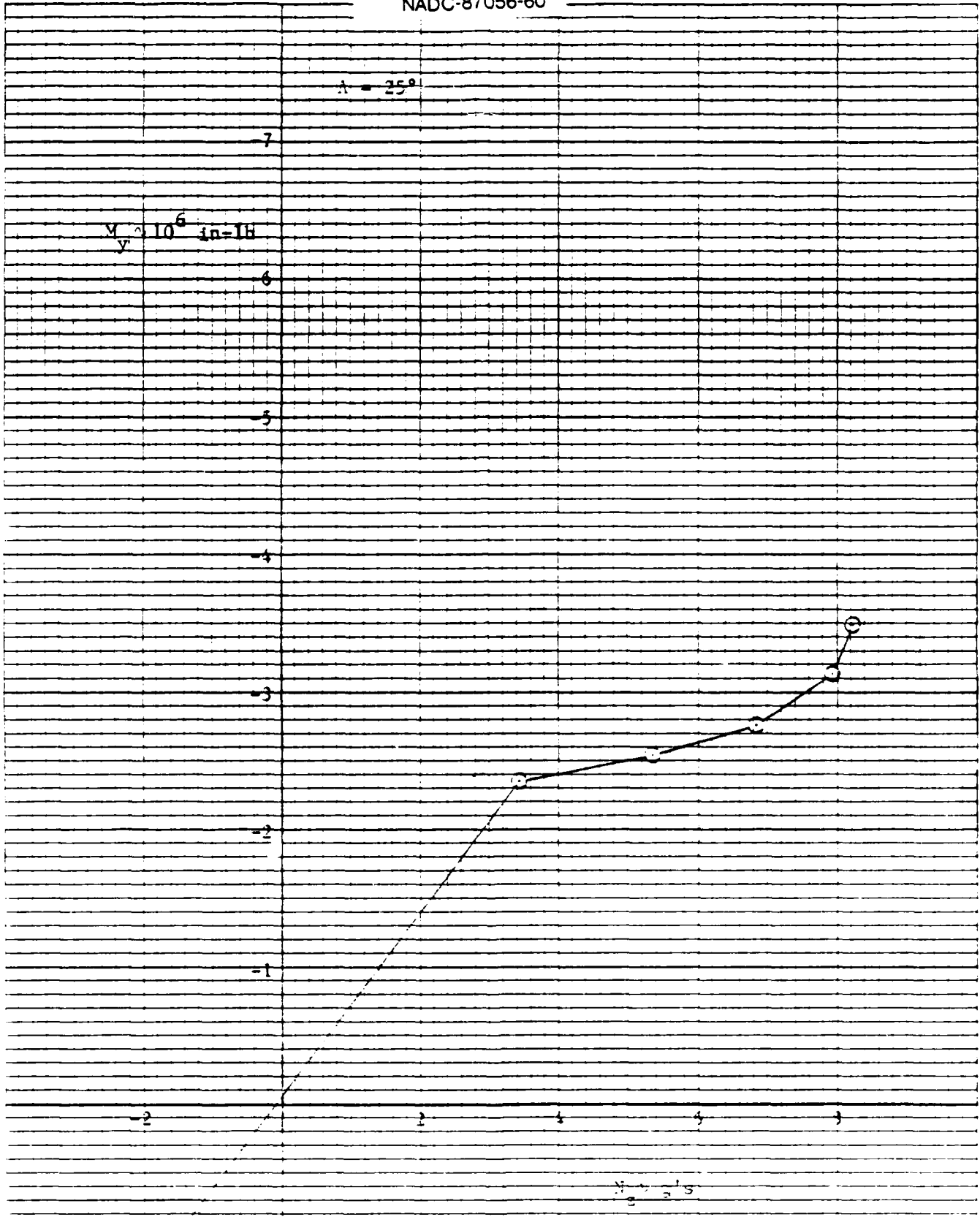


Figure 1. Net Wing Pivot Loads at 25° Sweep

1. Moment About y Axis



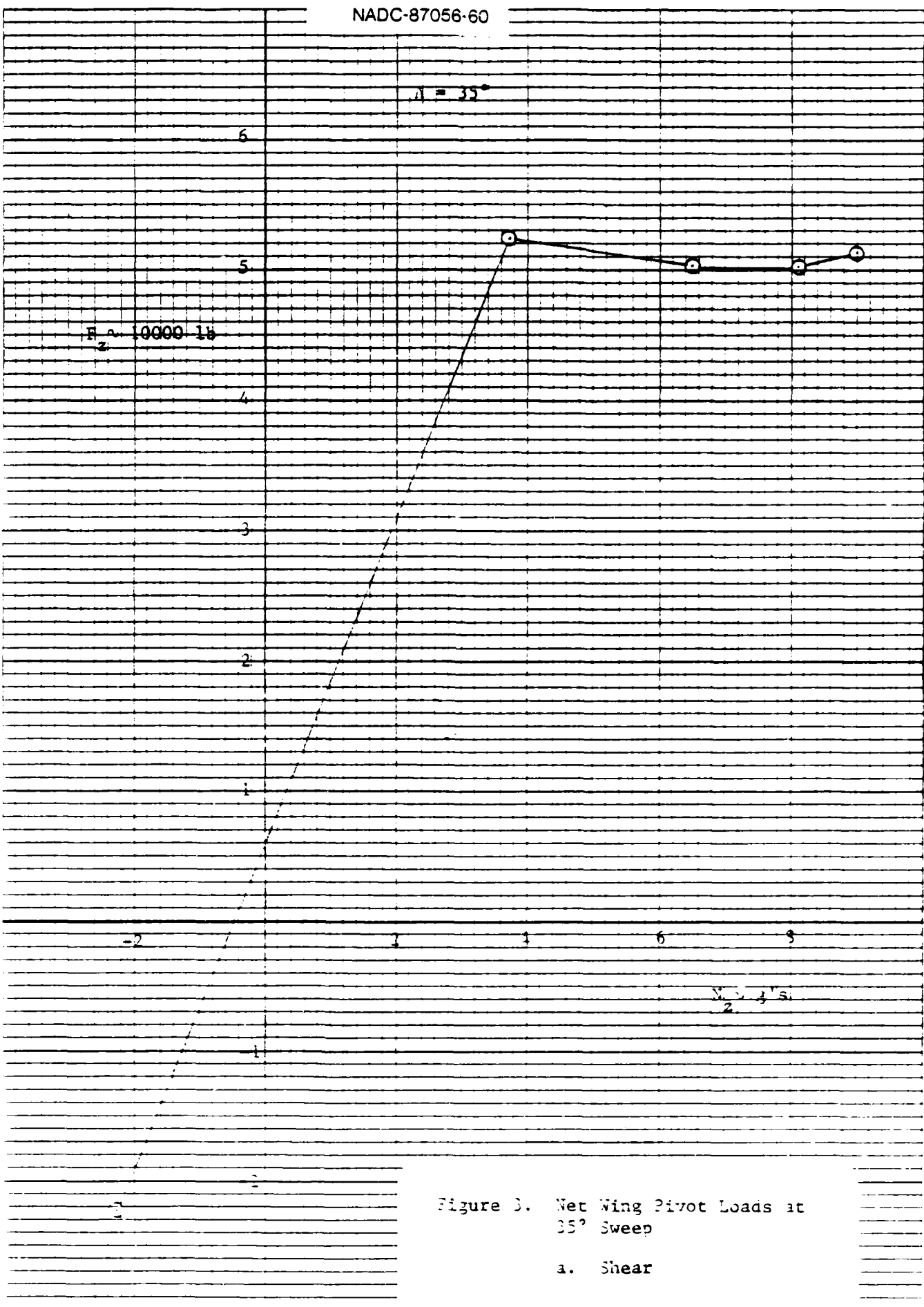


Figure 3. Net Wing Pivot Loads at 35° Sweep  
1. Shear

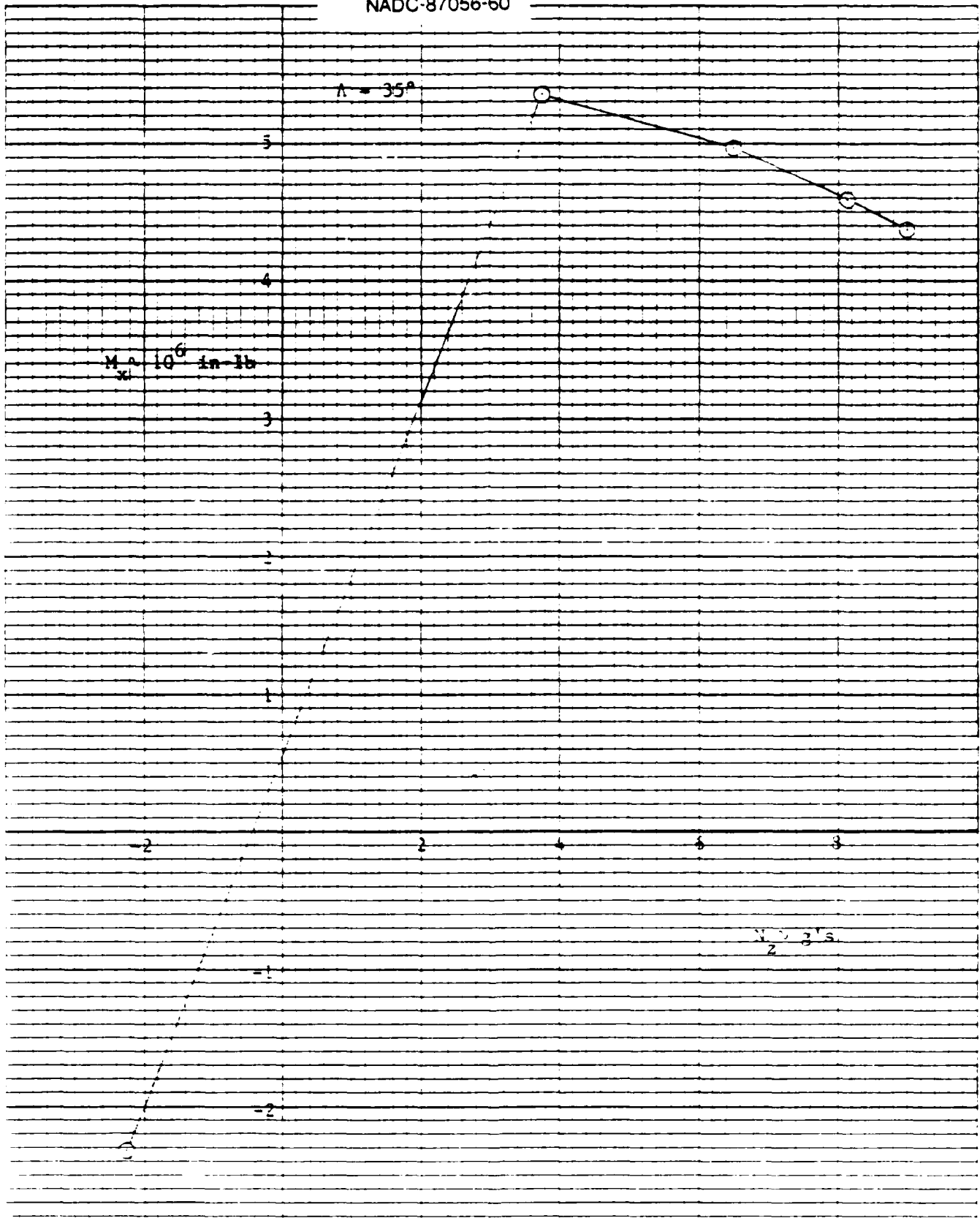


Figure 3. Net Wing Pivot Loads at 35° Sweep

5. Moment About x Axis

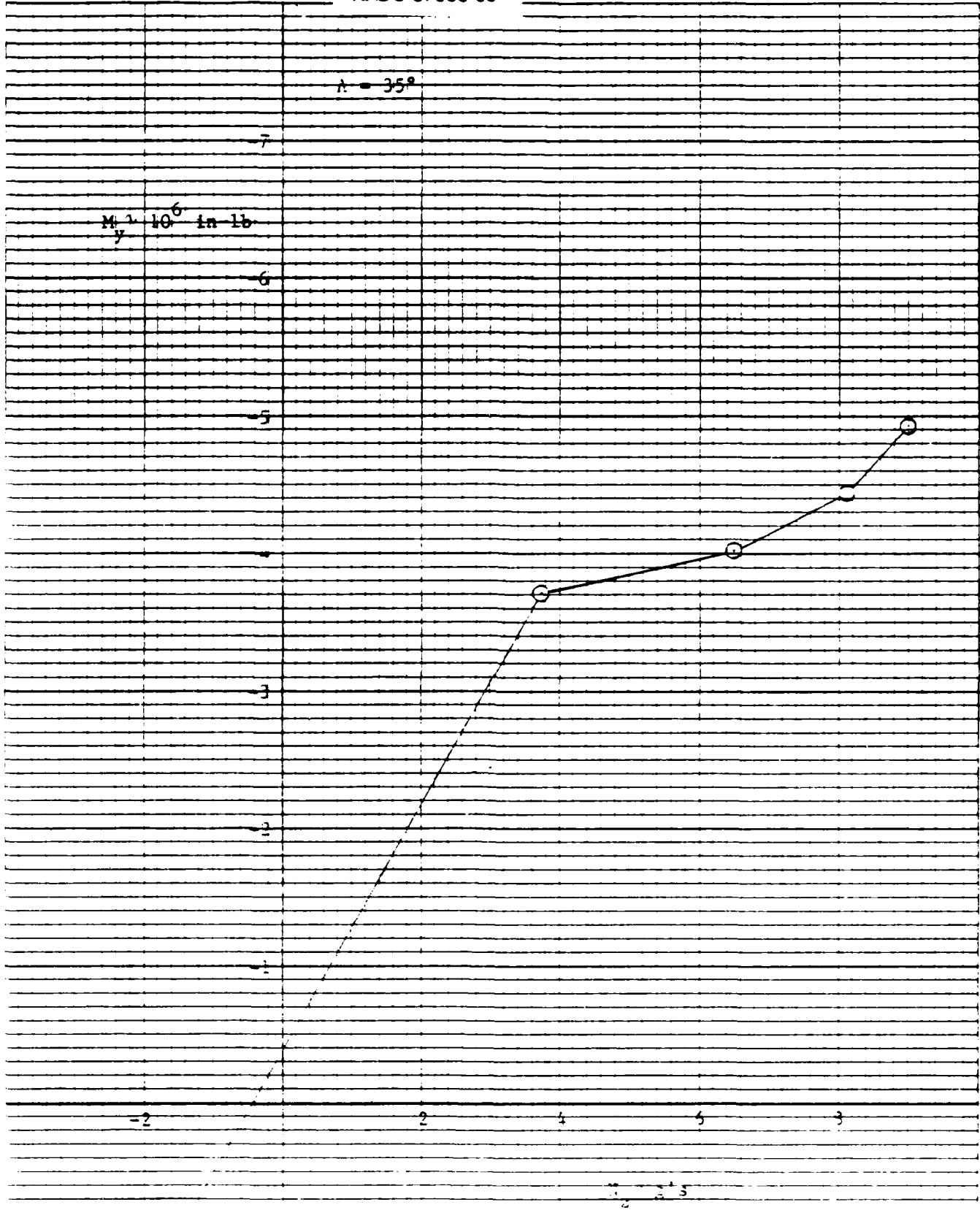


Figure 3. Net Wing Pivot Loads at  $35^\circ$  Sweep

1. Moment About Y AXIS

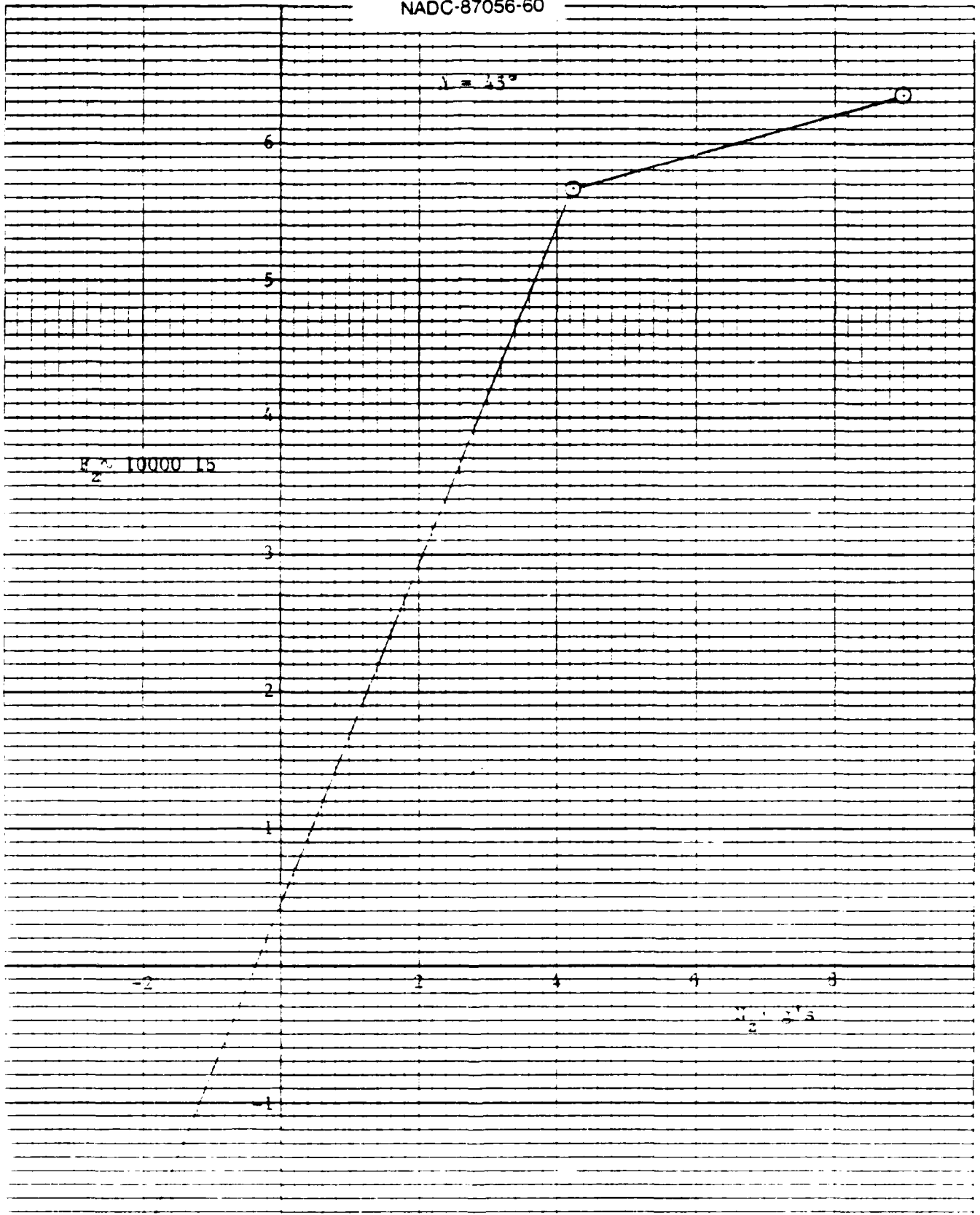


Figure 1. Net Wing Pivot Loads at 45° Sweep

1. Shear

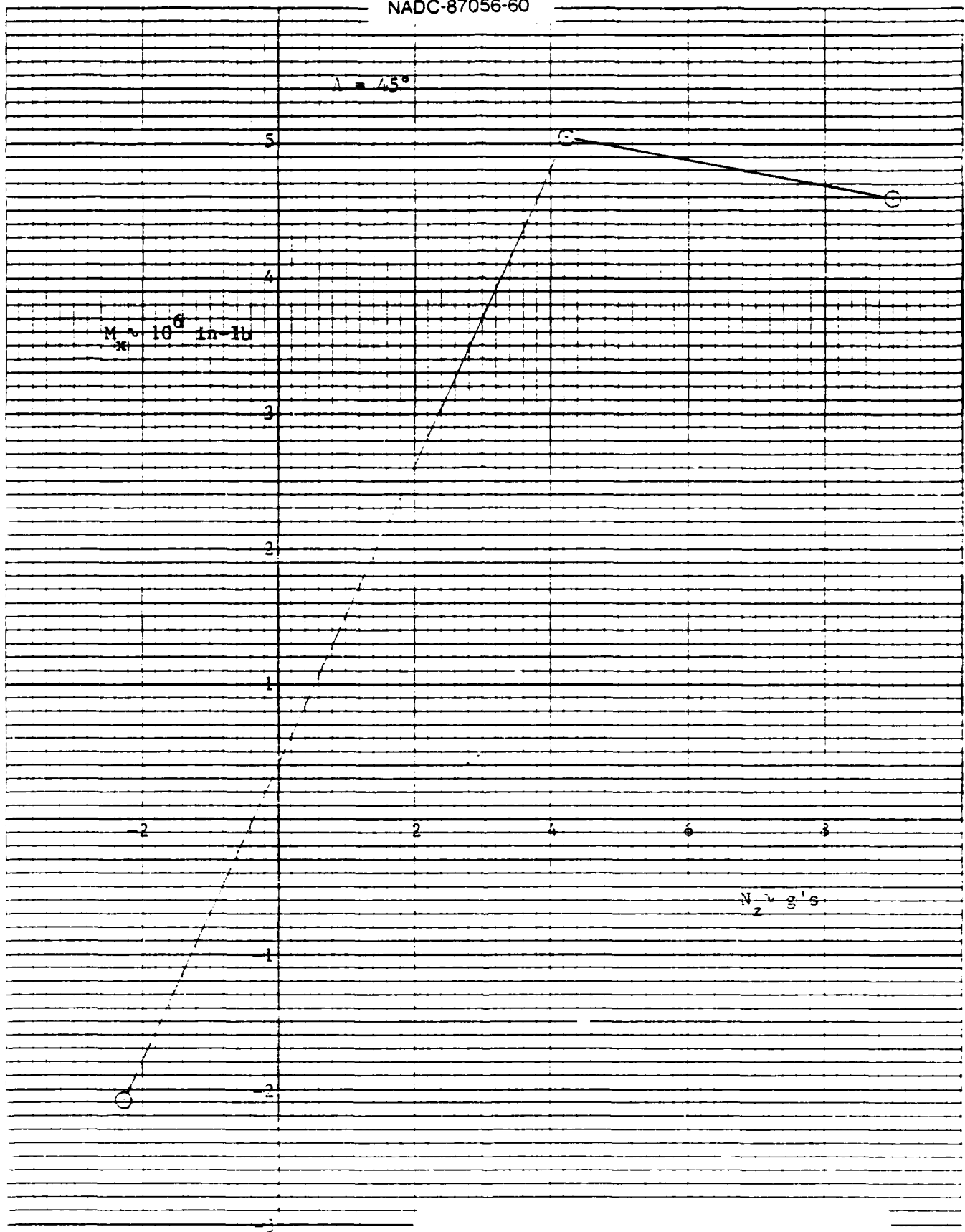


Figure 4. Net Wing Pivot Loads at 45° Sweep

5. Moment About x Axis

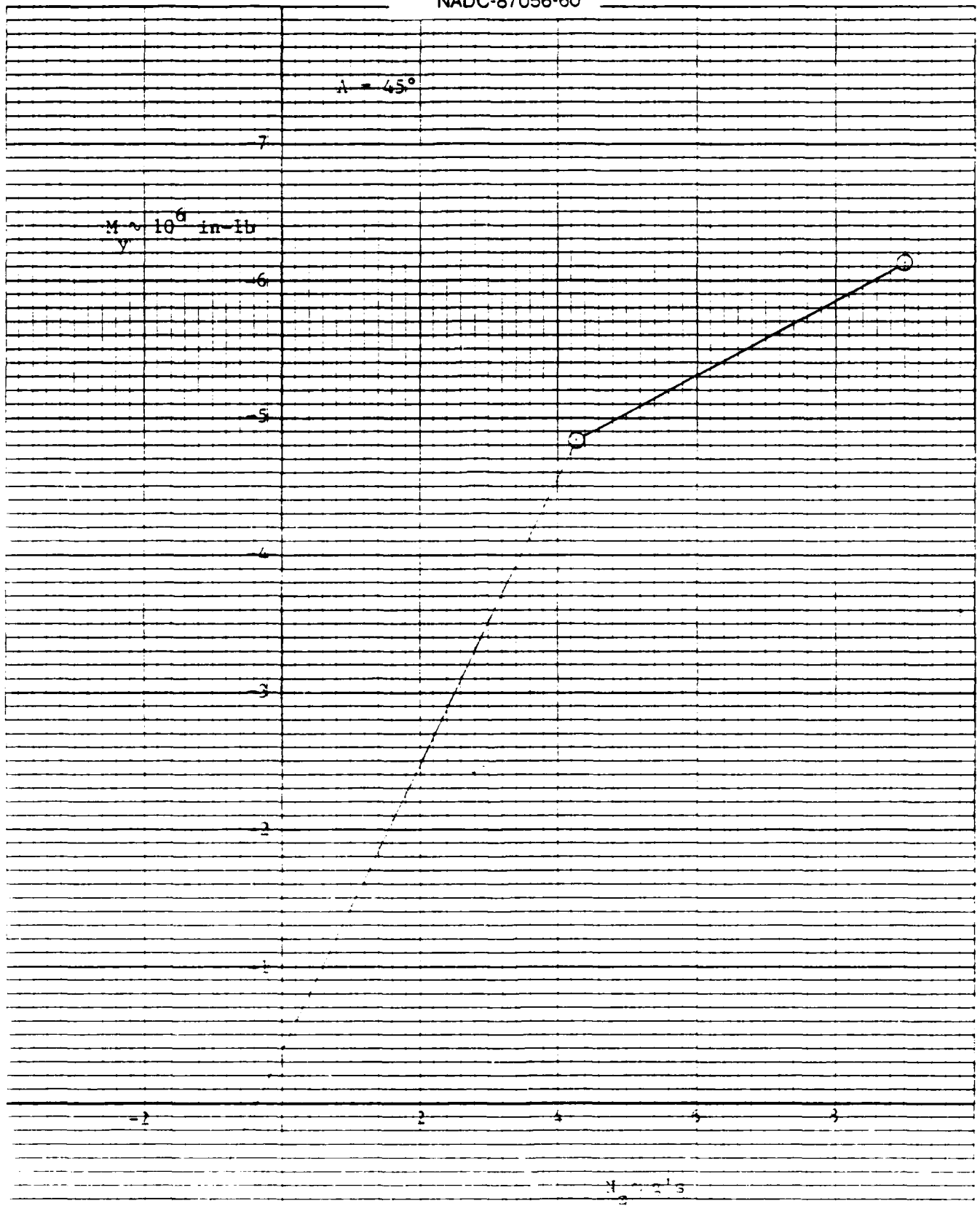


Figure 4. Net Wing Pivot Loads at 45° Sweep

1. Moment About y Axis

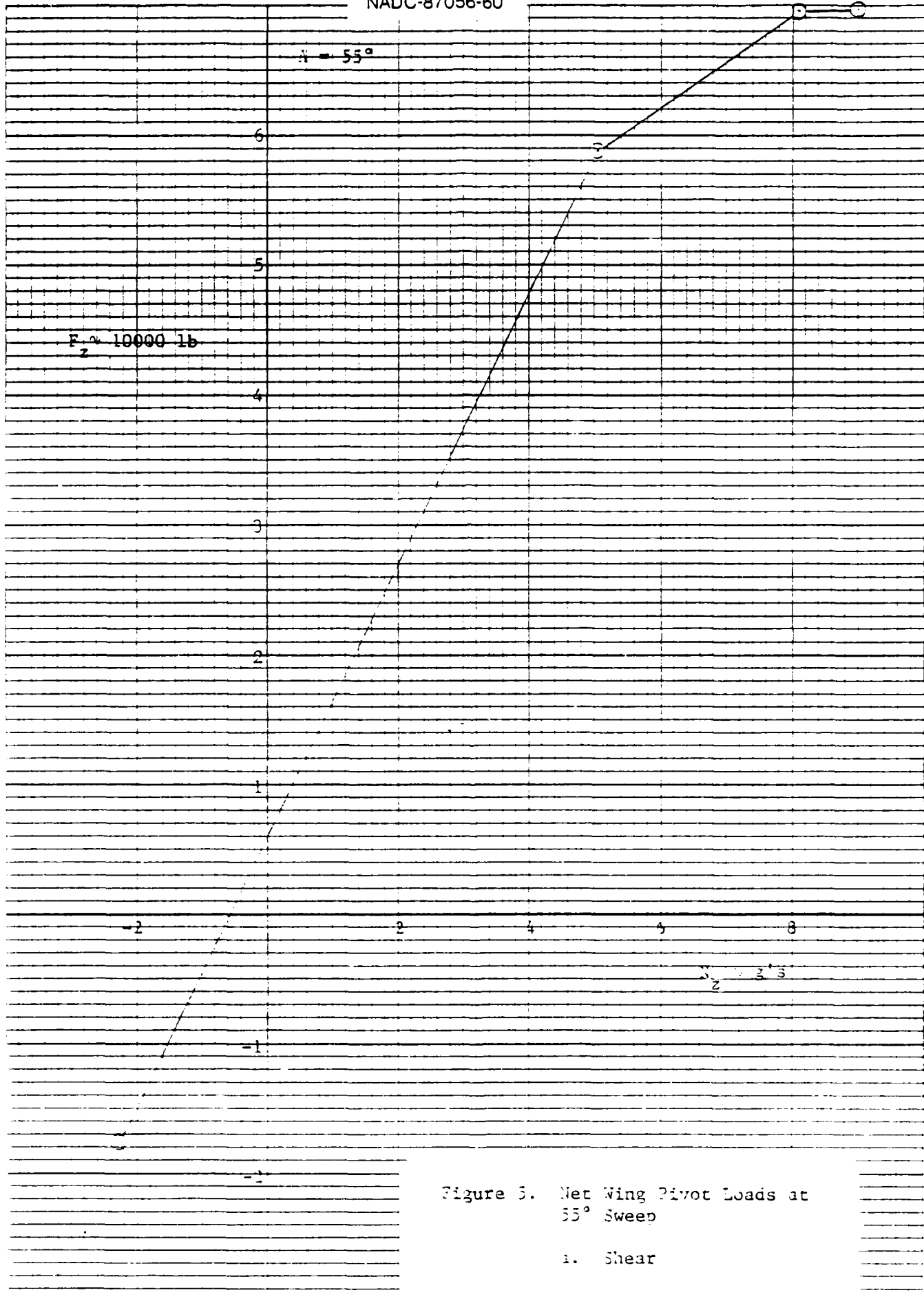


Figure 3. Net Wing Pivot Loads at 55° Sweep

1. Shear

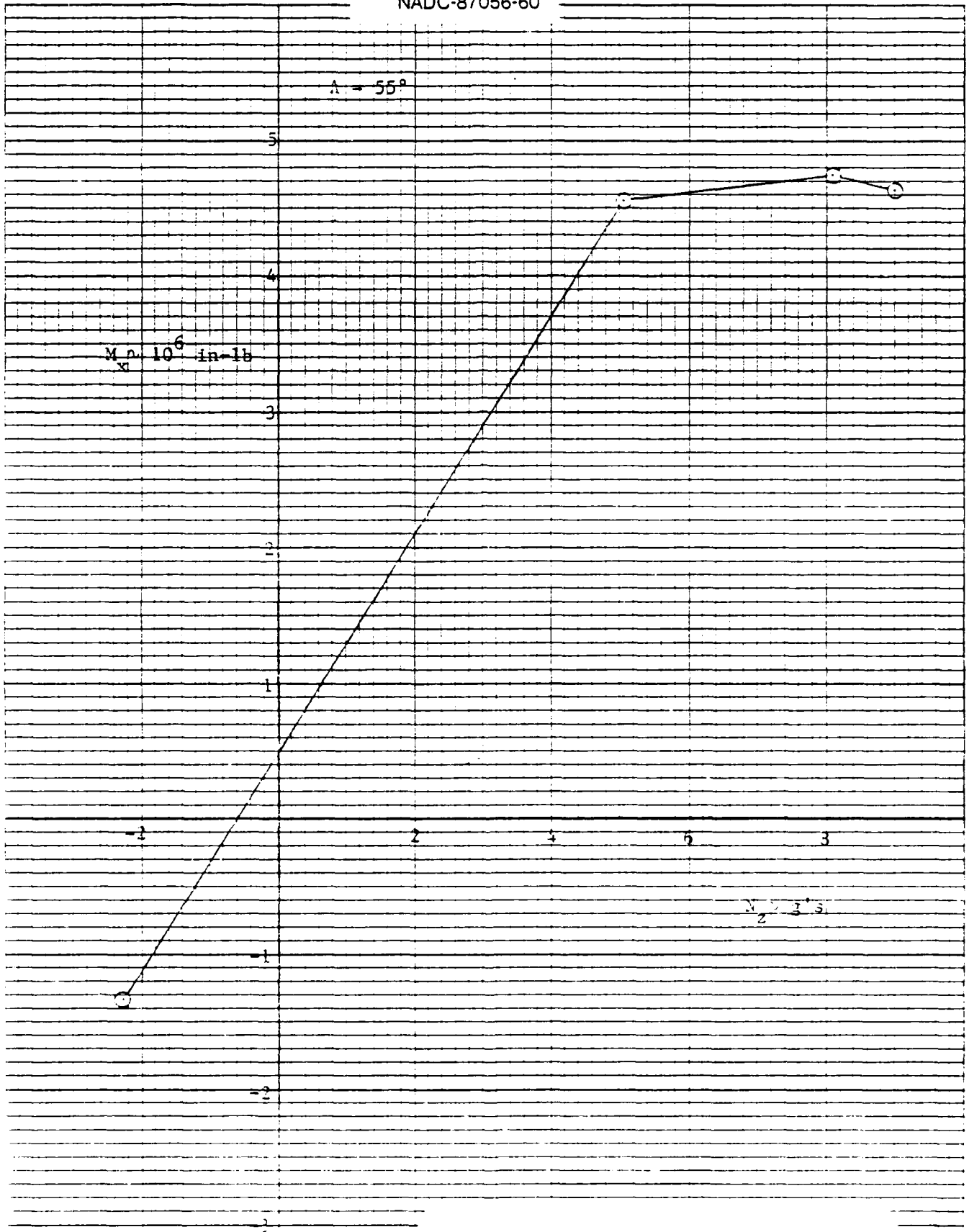


Figure 5. Net Wing Pivot Loads at 55° Sweep  
5. Moment About x Axis



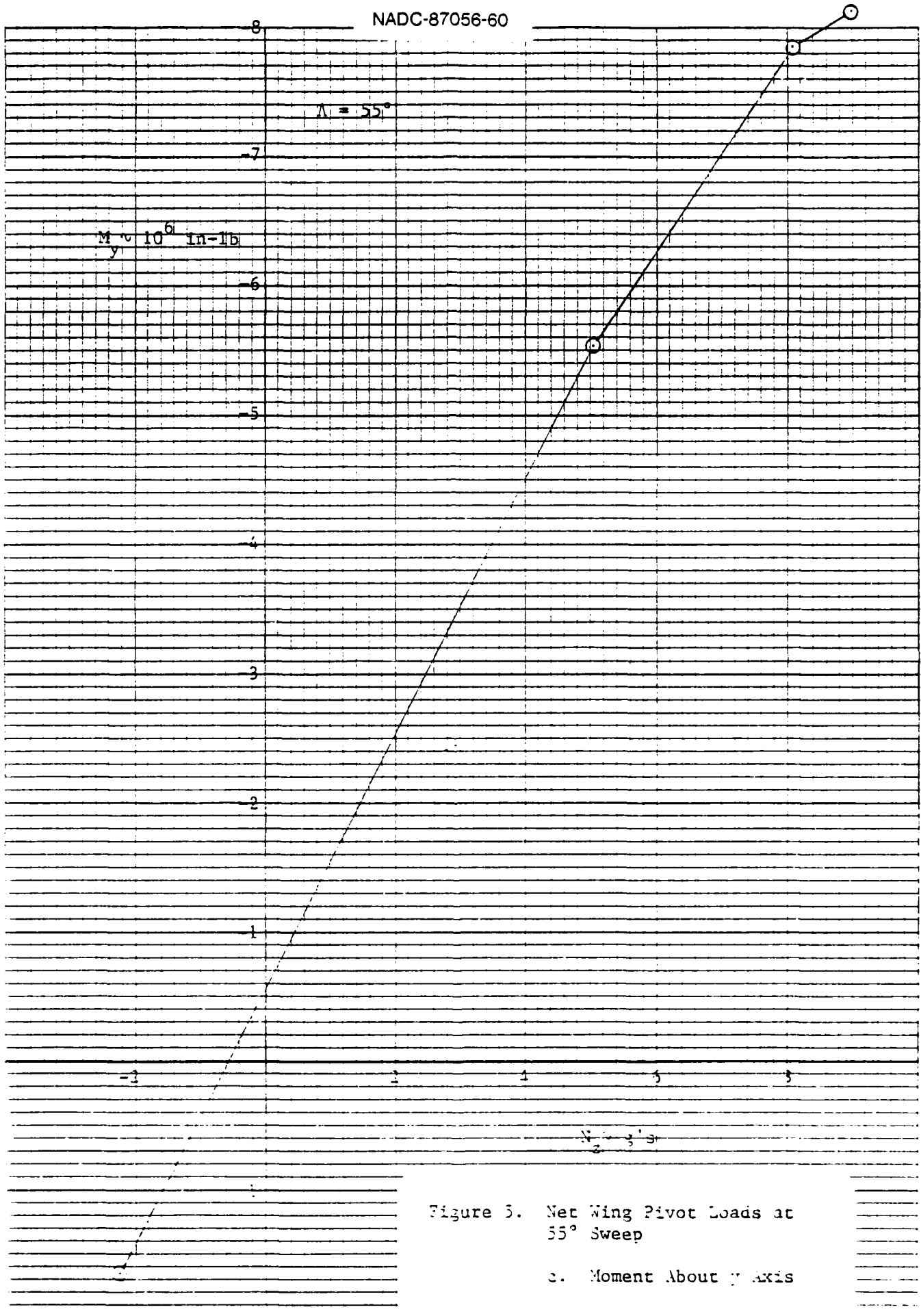


Figure 5. Net Wing Pivot Loads at 55° Sweep

c. Moment About Y Axis

$\Lambda = 68^\circ$

$F_z = 10000 \text{ lb}$



Figure 6. Net Wing Pivot Loads at  $68^\circ$  Sweep

1. Shear

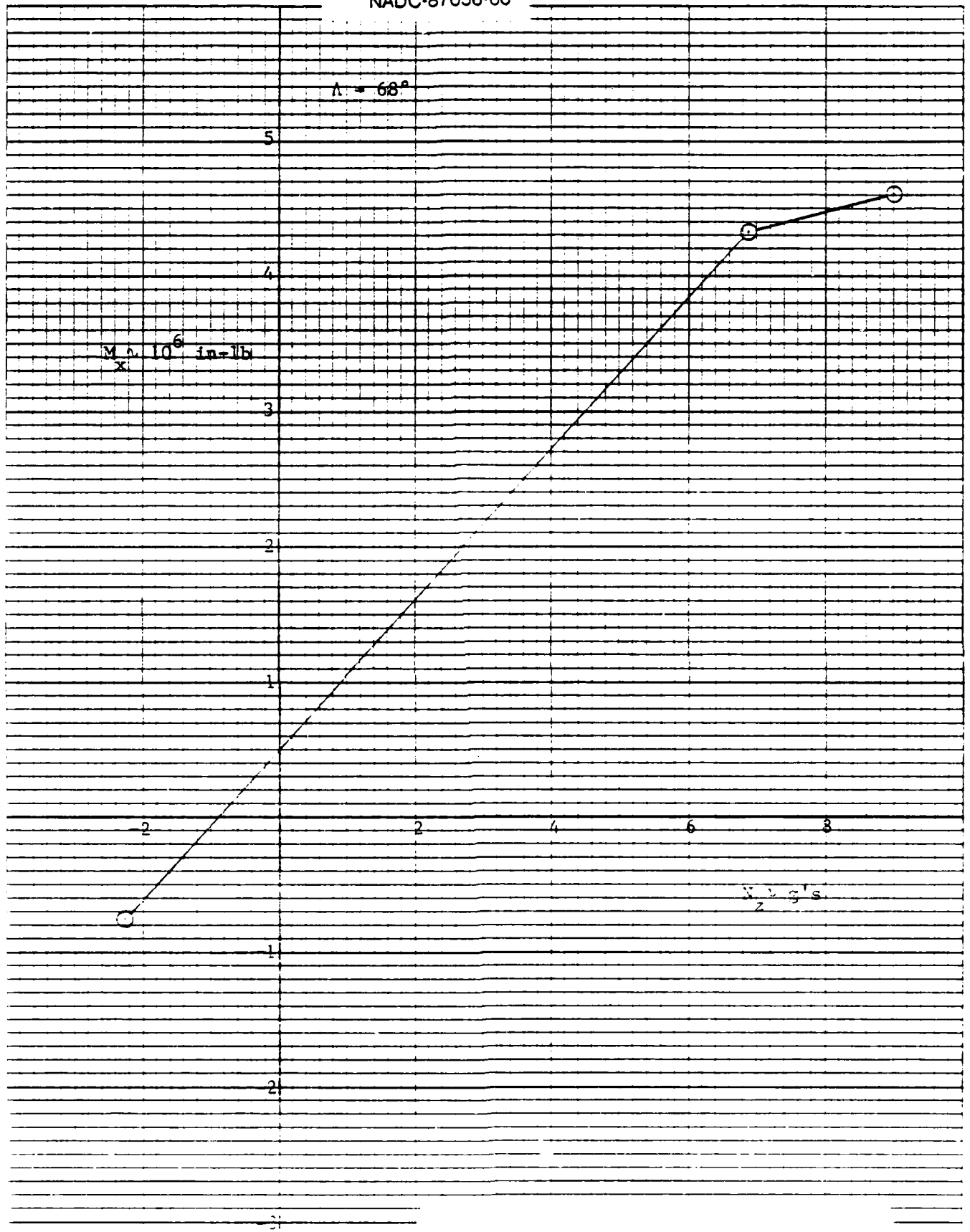


Figure 6. Net Wing Pivot Loads at 68° Sweep

6. Moment About x Axis

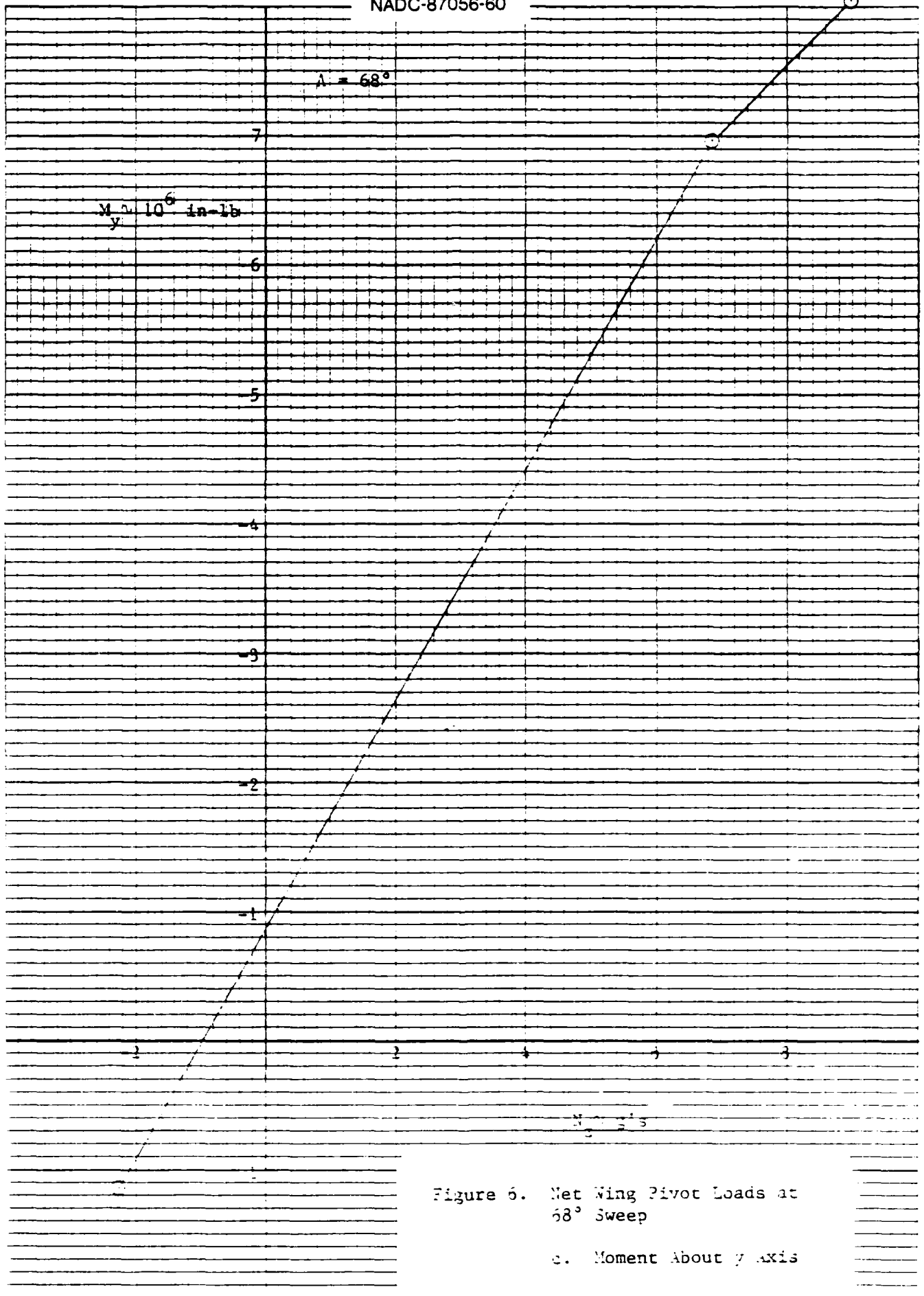


Figure 6. Net Wing Pivot Loads at  $68^\circ$  Sweep

a. Moment About y Axis

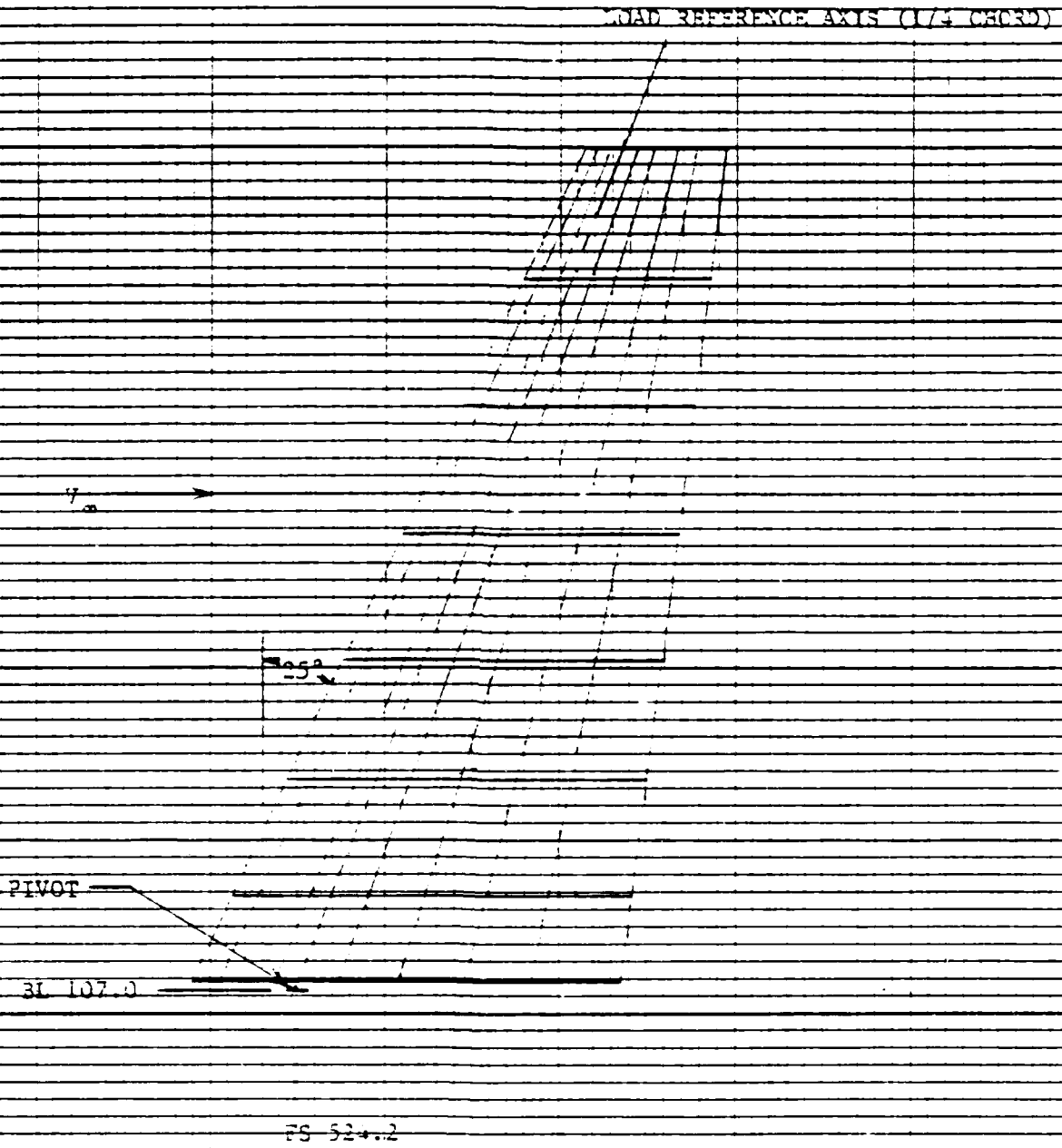


Figure 7. Aerodynamic Model of F-14 Wing Outer Panel

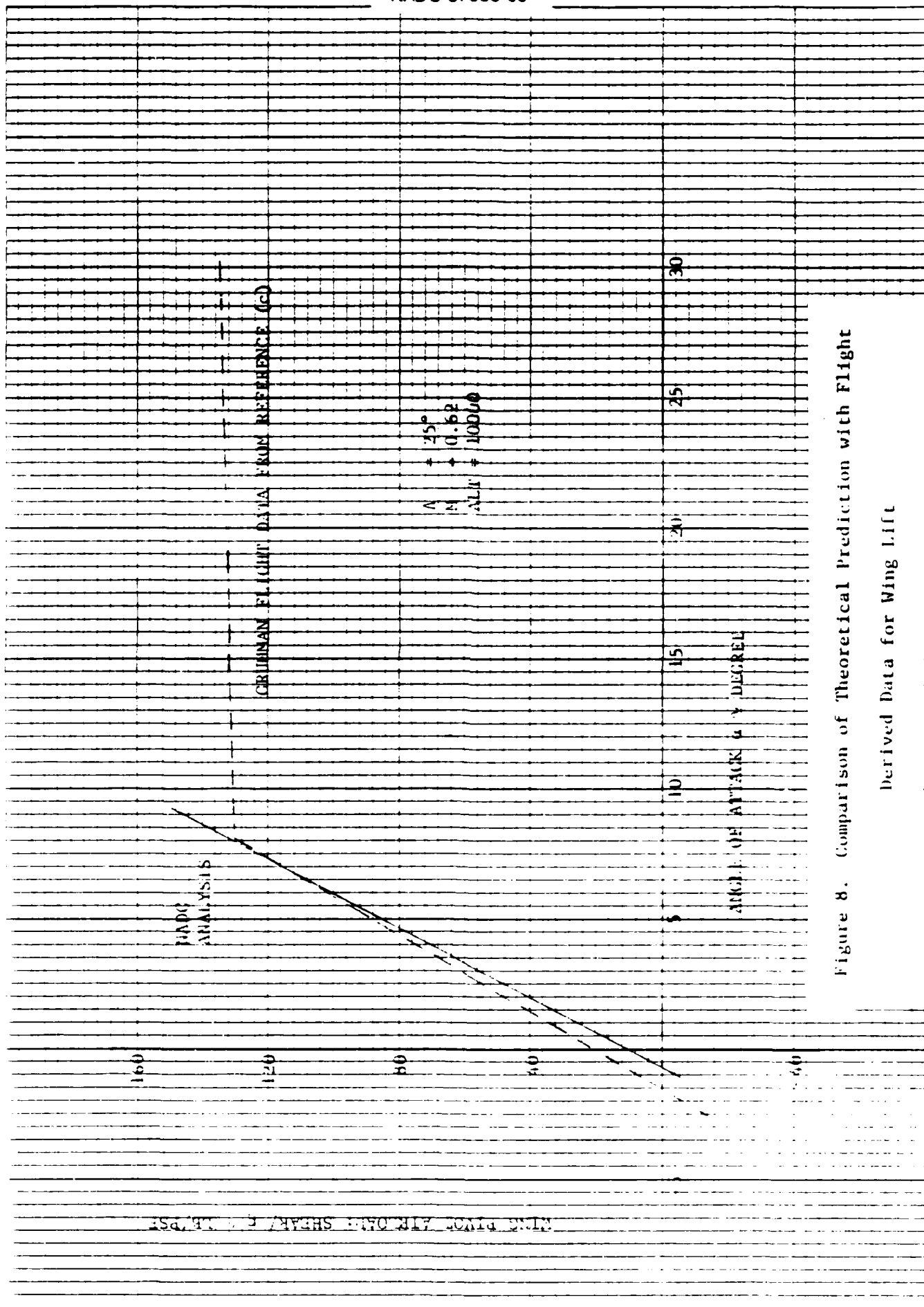


Figure 8. Comparison of Theoretical Prediction with Flight  
 Derived Data for Wing Lift

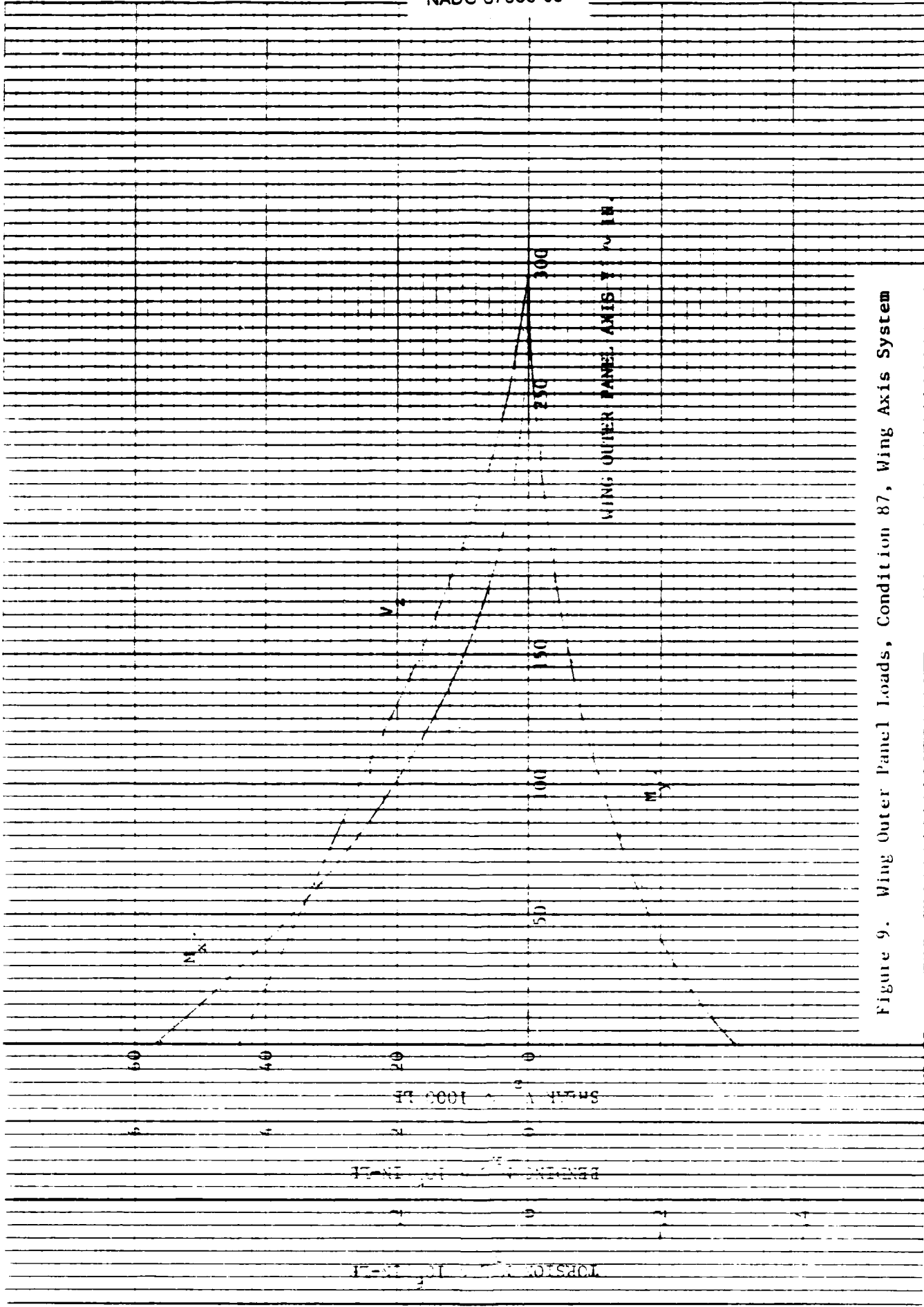


Figure 9. Wing Outer Panel Loads, Condition 87, Wing Axis System

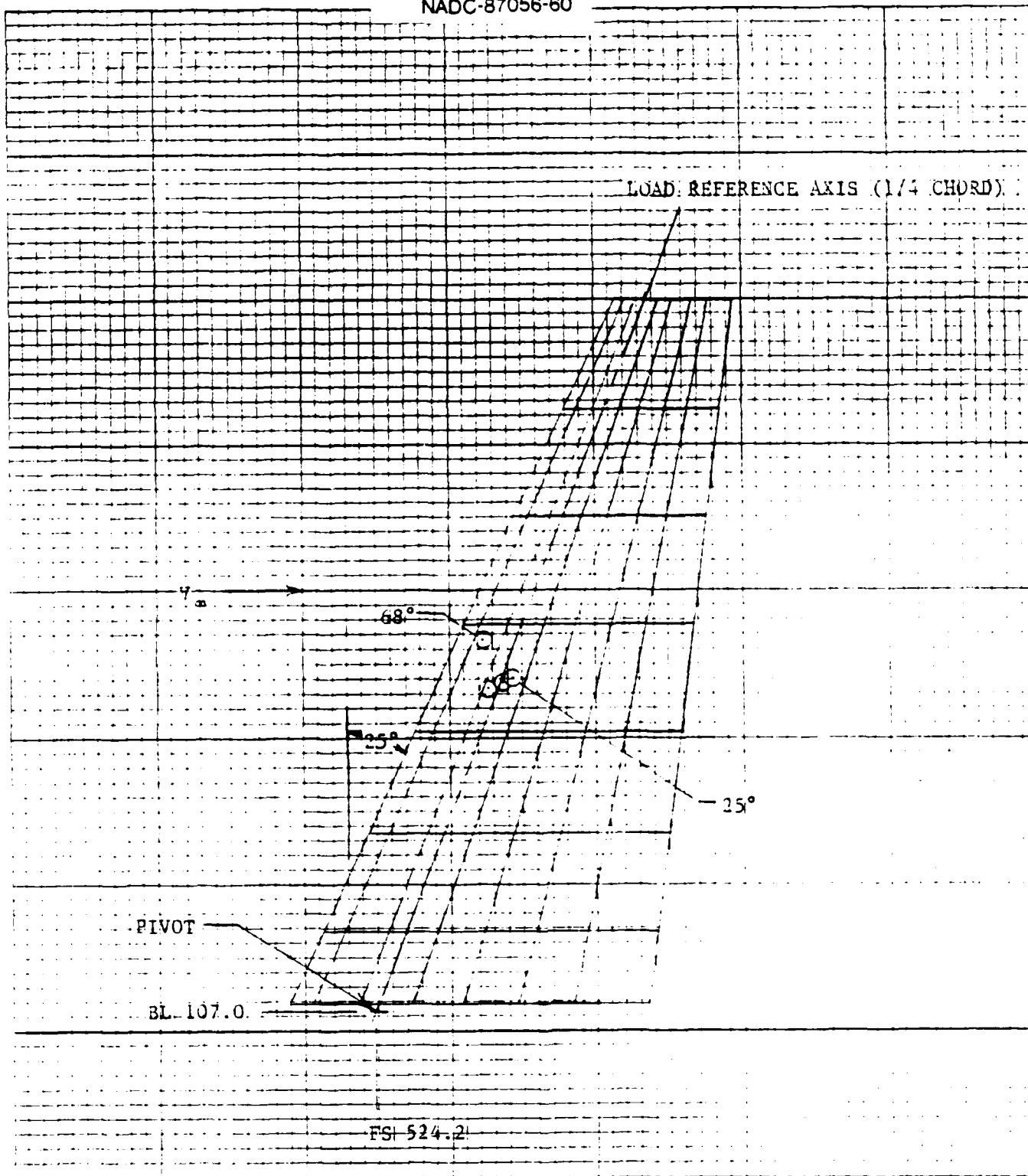


Figure 10. Wing Center of Pressure Location at Various Sweep Angles



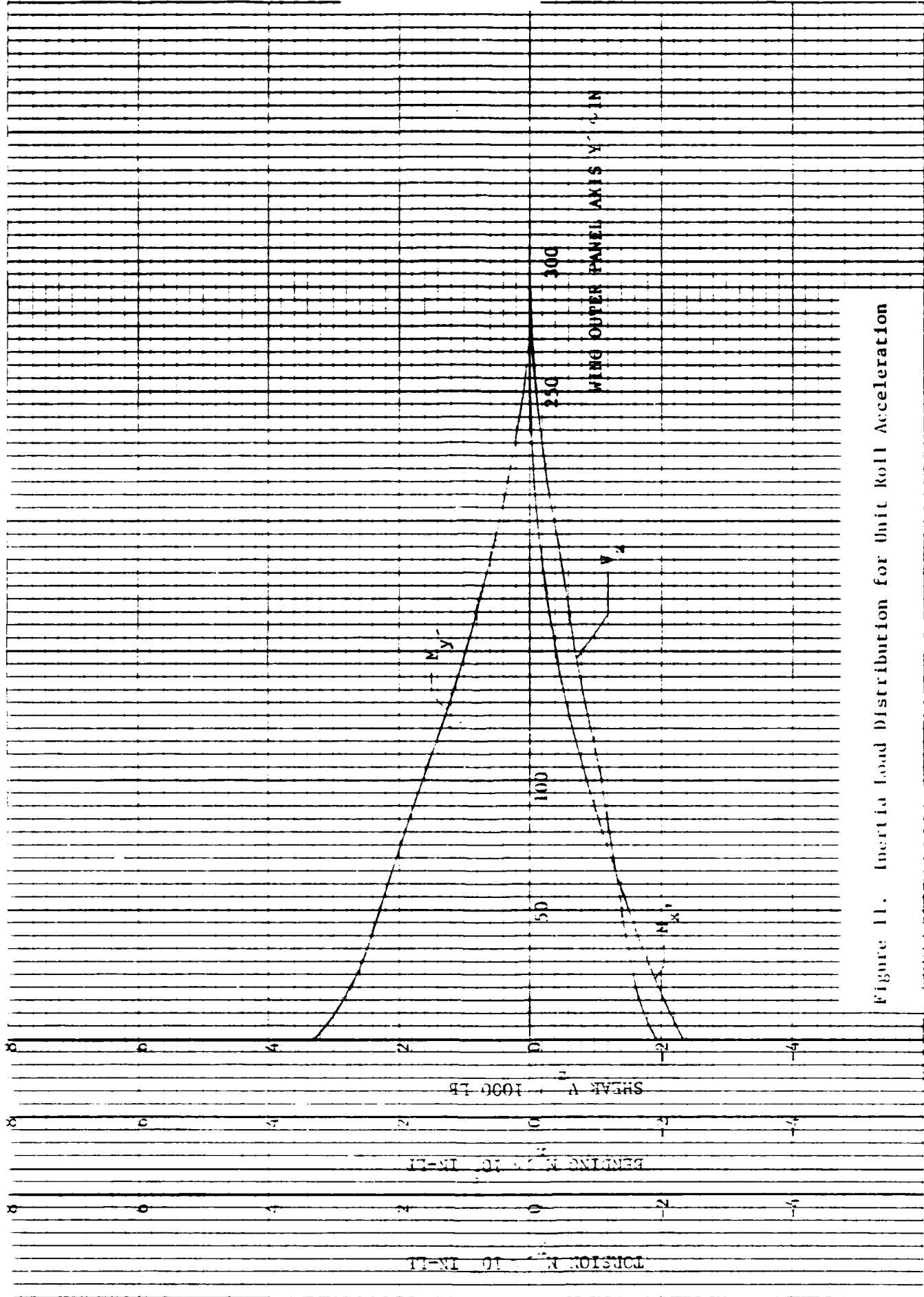


Figure 11. Inertia Load Distribution for Unit Roll Acceleration

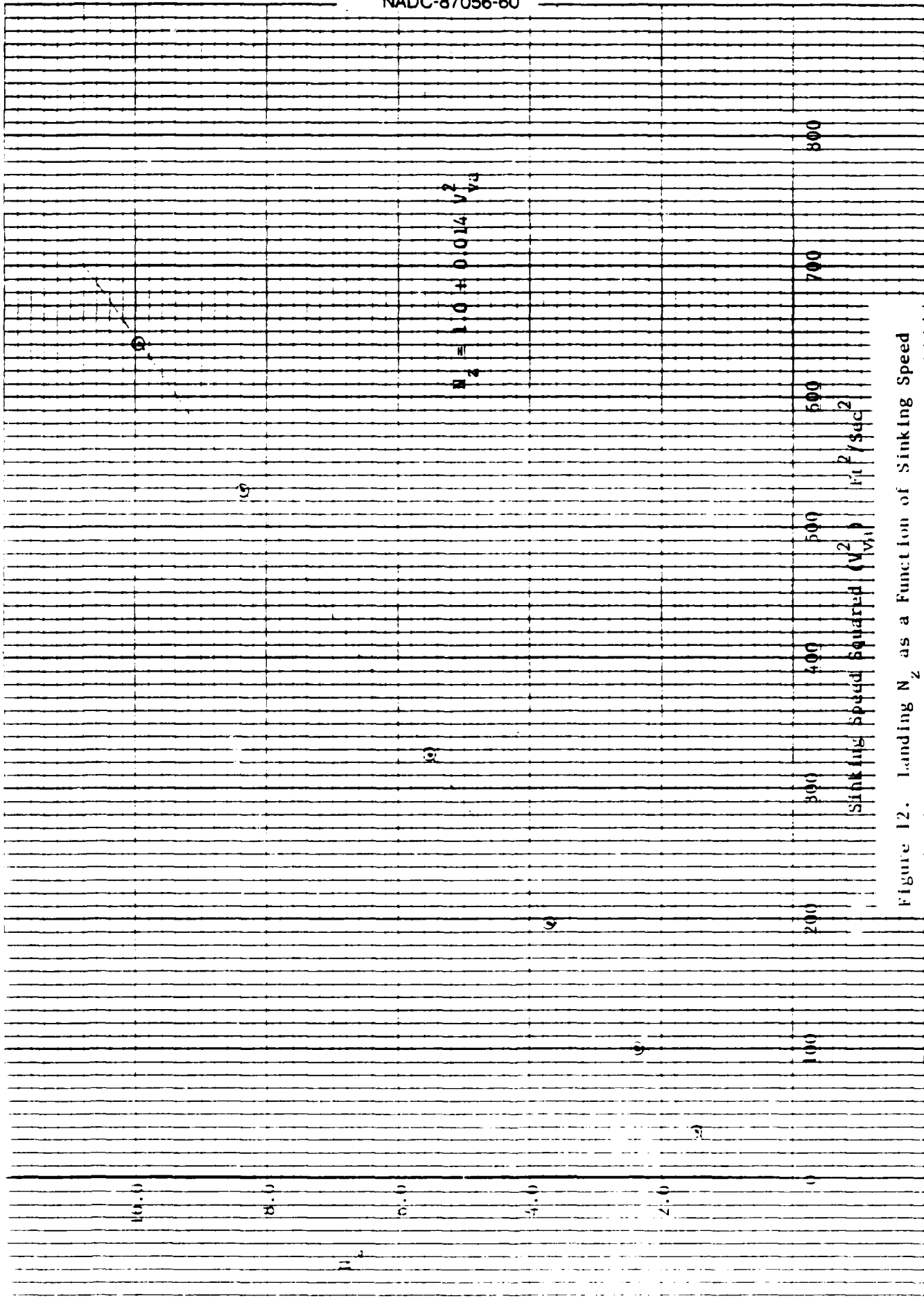


Figure 12. Landing  $N_z$  as a Function of Sinking Speed

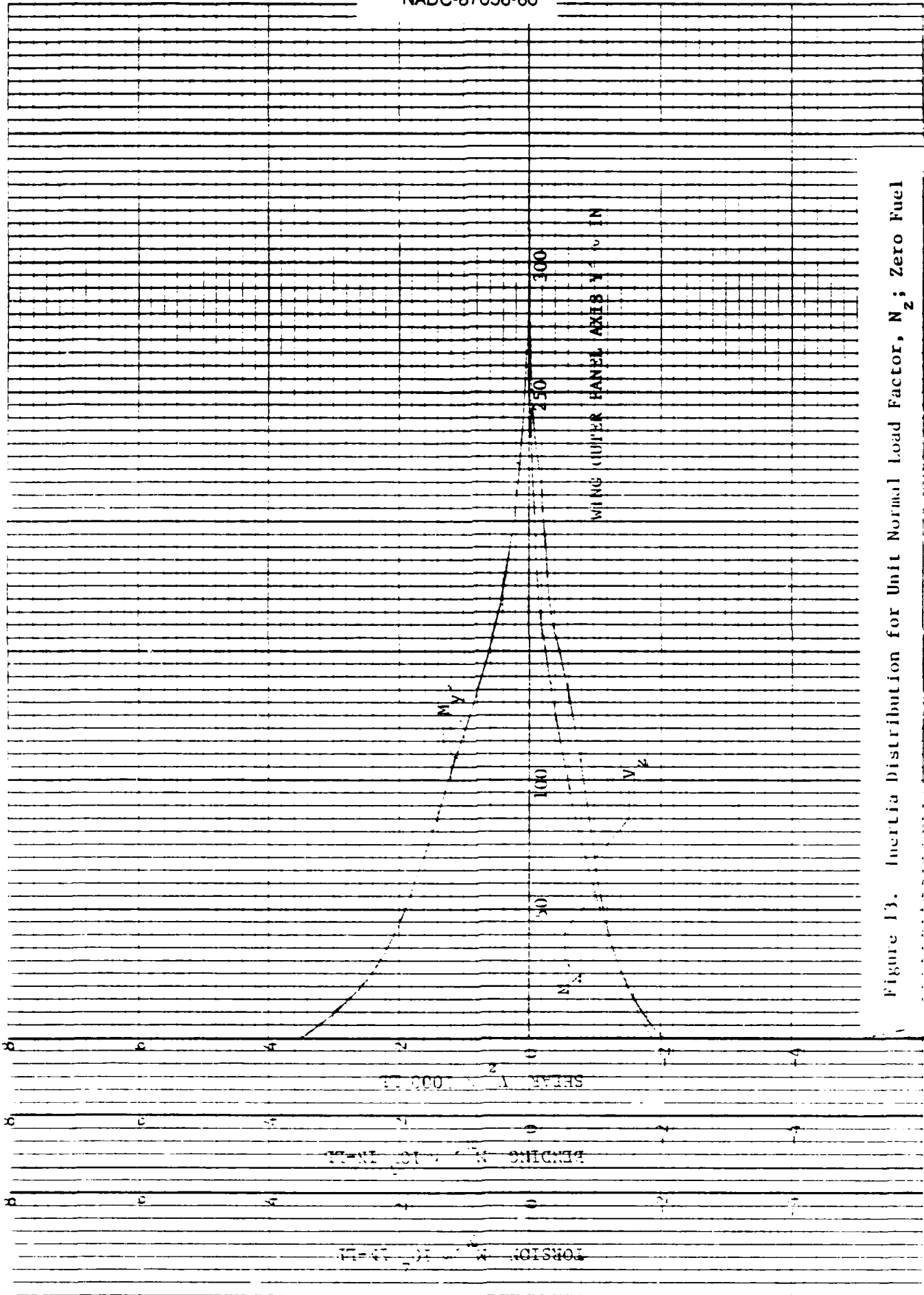


Figure 13. Inertia Distribution for Unit Normal Load Factor,  $N_2$ ; Zero Fuel

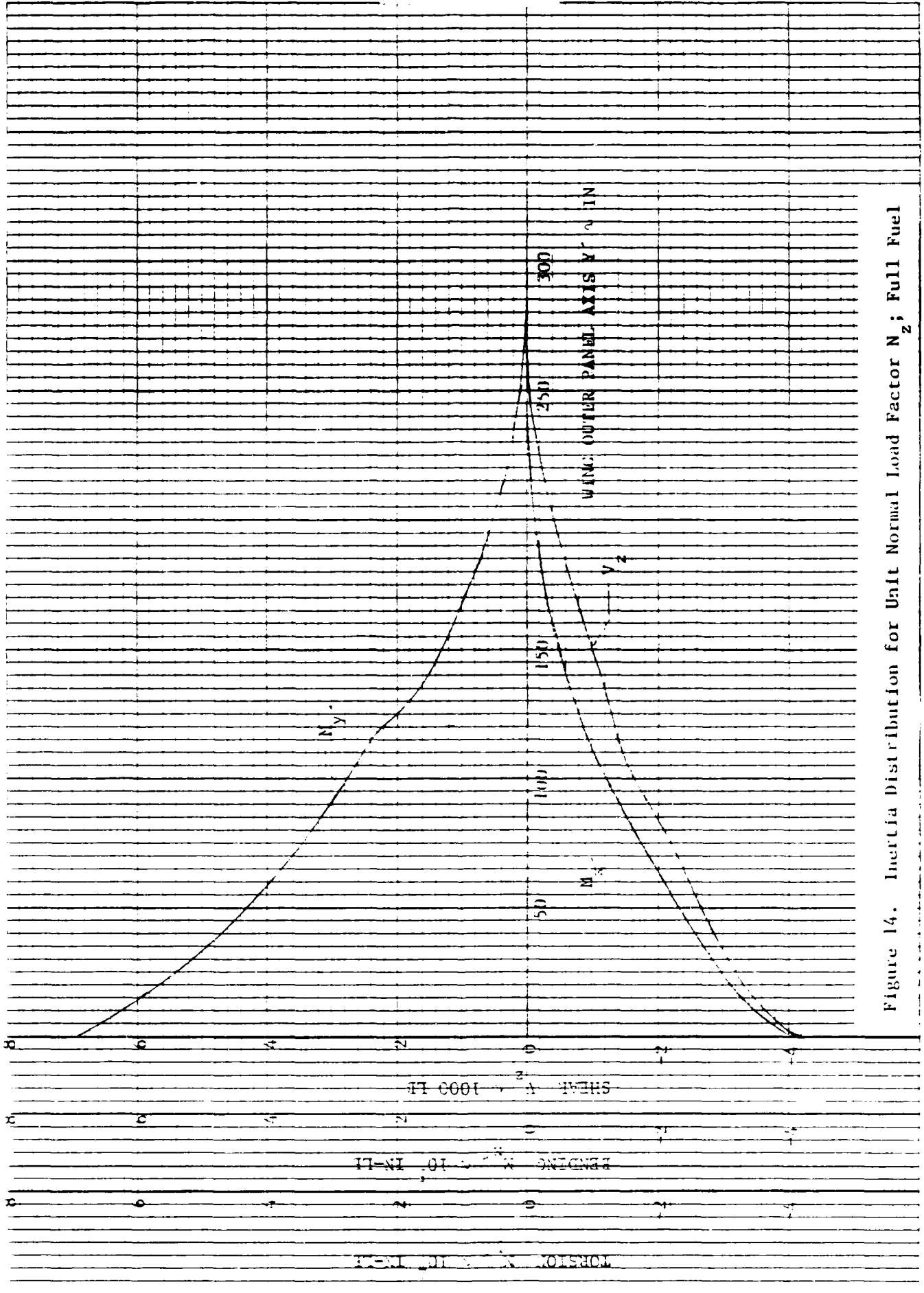


Figure 14. Inertia Distribution for Unit Normal Load Factor  $N_2$ ; Full Fuel

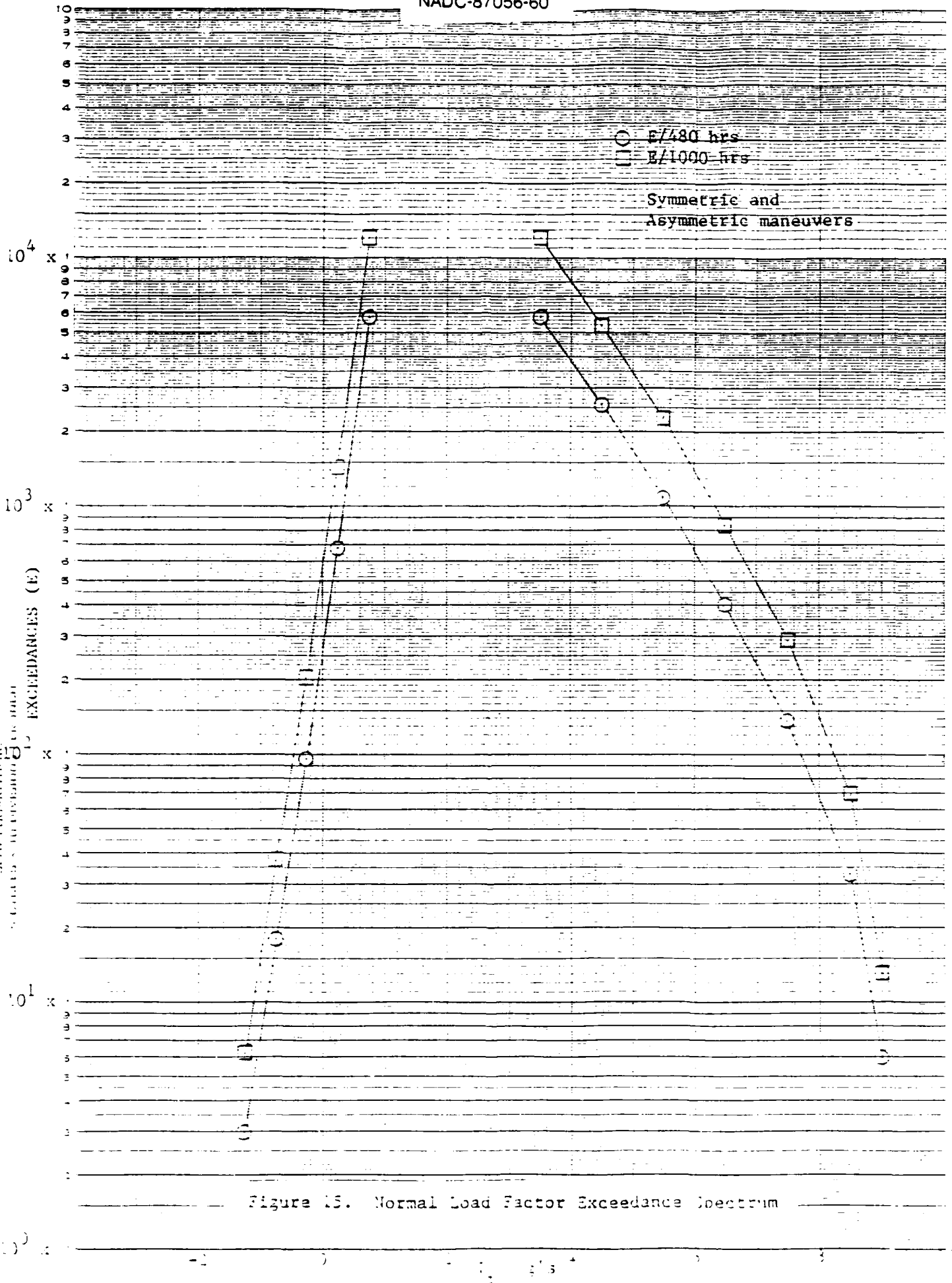
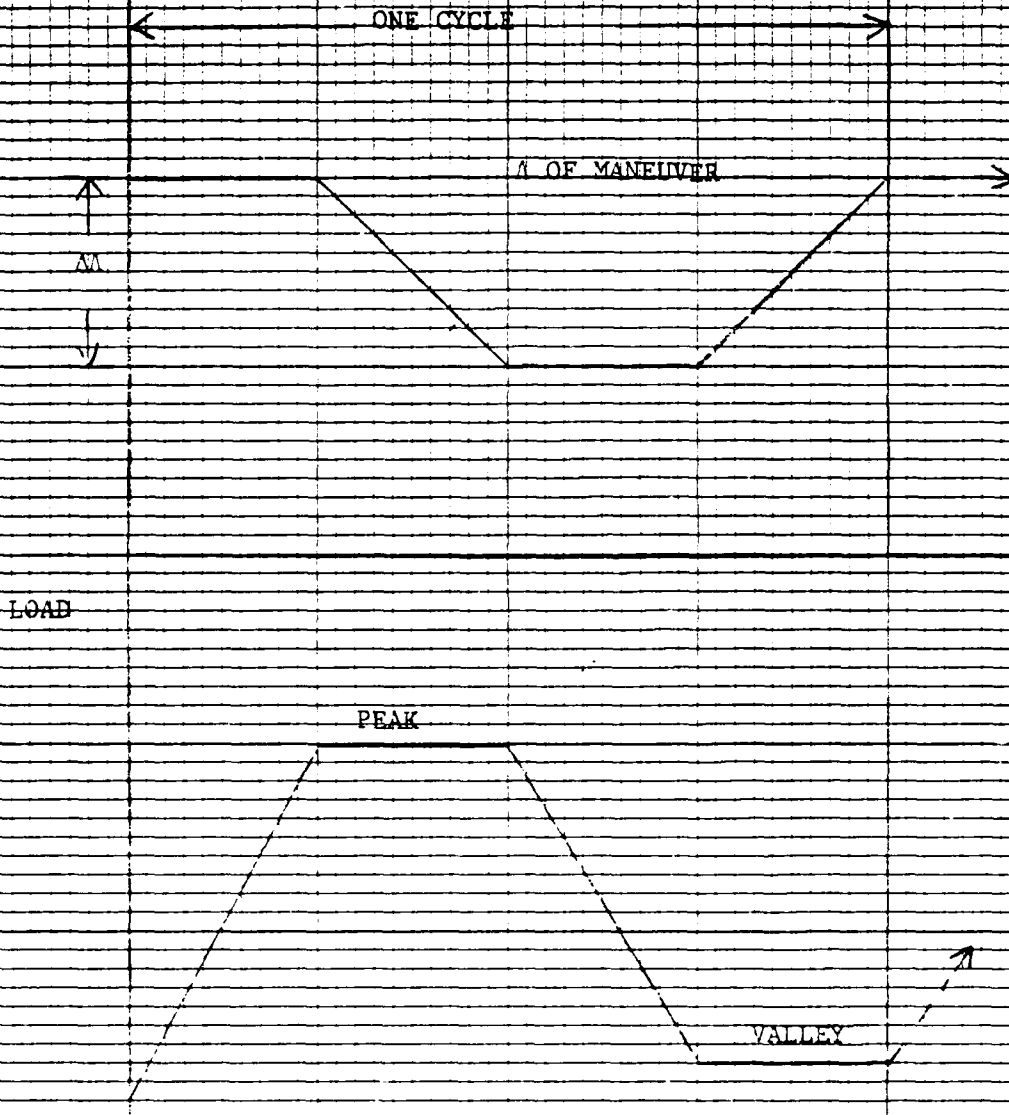


Figure 15. Normal Load Factor Exceedance Spectrum

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION  
 WASHINGTON, D. C. 20546-0001  
 REPORT NUMBER 87-056-60

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 DATE OF REVISION 1987



Note:  $\Delta = 5^\circ$ , if  $\Delta_{LE} = 25^\circ$  or if  
 bearing load  $\geq 650000$  lb  
 $\Delta = 10^\circ$  for any other  $\Delta_{LE}$  and if  
 bearing load  $< 650000$  lb

Figure 16. Wing Pivot Cycle

APPENDIX (A)

SEQUENTIAL SPECTRUM FOR AIRCRAFT #98 TEST

NOTE: Appendices (A) and (B) contain a flight-by-flight random sequence of load levels for 440 flights (480 hours). All printout lines (records) have the same format, which is described below:

<u>Columns</u>	<u>Format</u>	<u>Description</u>
1-4	I4	Flight number identification (1 to 440)
5-7	3X	Not used
8-9	I2	The number of loads in this line
10-69	20I3	These numbers identify the load conditions to be applied to the test article.

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1 20156156 87 87159155 87 87157155 156156156156255256157155156156  
1 20 87 87157155 87 87157155 156156156255256157155156156 87 87  
1 20159155 87 87158155 156156229230156156255256157154156156 87 87  
1 16157155 156156156156255256157155156156 J8 39156156  
2 20156156 87 87157155 87 87157155 156156227228156156255256157155  
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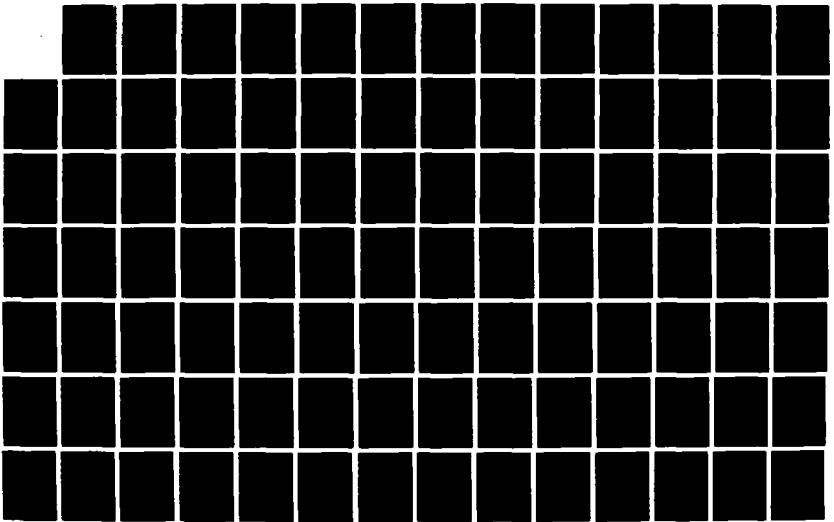
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F-14 WING OUTER PANEL FATIGUE TEST SPECTRUM(U) NAVAL  
AIR DEVELOPMENT CENTER WARMINSTER PA AIR VEHICLE AND  
CREE SYSTEMS TECHN OLOGY DIRECTORATE G S SEIDEL ET AL.  
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MICROCOPY RESOLUTION TEST CHART  
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425 4 15 16156156  
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433 2156156

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440 2014614715615656571561563334358515615648495051156156  
440 613141516156156

NADC-87056-60

APPENDIX (B)

SEQUENTIAL SPECTRUM FOR WING OUTER PANEL TEST

1 15156 87157155 87157155156227228156273274160155 0 0 0 0 0 0  
1 13156 87159155 87157155156255256157155156 0 0 0 0 0 0  
1 13 87157155 87157155156255256157155156 87 0 0 0 0 0 0  
1 15159155 87158155156229230156255256157154156 87 0 0 0 0 0 0  
1 10157155156255256157155156 20156 0 0 0 0 0 0 0 0  
2 15156 87157155 87157155156227228156255256157155 0 0 0 0 0 0  
2 13156 87158155 87157155156263264158155156 0 0 0 0 0 0  
2 15 87158155 87159155156227228156255256157155156 0 0 0 0 0 0  
2 13 87157155 87159155156255256157154156 87 0 0 0 0 0 0  
2 11157155156255256157155156 17156 9 0 0 0 0 0 0 0  
2 10 85156 20156 9 85156 20156 20 0 0 0 0 0 0 0 0  
2 2 85156 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0  
3 13156 87157155 87157155156261262158155156 0 0 0 0 0 0  
3 13 87158155 87157155156273274160154156 87 0 0 0 0 0 0  
3 14157152 87158155156261262158155156 87157155 0 0 0 0 0 0  
3 12 87157155156255256157155156 87160155 0 0 0 0 0 0  
3 8156261262158155156 20156 0 0 0 0 0 0 0 0 0  
4 13156 87157155 87157155156263264158155156 0 0 0 0 0 0  
4 13 87158153 87157155156275276160155156 87 0 0 0 0 0 0  
4 16158155 87157152156227228229230156255256157155156 0 0 0 0  
4 15 87158155 87157155156229230156255256157155156 0 0 0 0  
4 13 87157155156229230156267268159155156 20 0 0 0 0 0 0  
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5 15 87157155 87157155156227228156261262158155156 0 0 0 0  
5 13 87157155 87157155156255256157155156 87 0 0 0 0 0 0  
5 10159154156273274160155156 20156 0 0 0 0 0 0 0 0  
6 15156 87157155 87158155156229230156255256157155 0 0 0 0  
6 13156 87157155 87157155156267268159154156 0 0 0 0 0 0  
6 13 87157155 87161155156255256157154156 87 0 0 0 0 0 0  
6 14161155 87158155156267268159155156 87159155 0 0 0 0 0 0

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6	3 84	74156	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	
7	13150	87157155	87 91155156255256157154156	0 0 0 0 0 0 0 0	
7	15	87159155	87158155156227228156261262158155156	0 0 0 0 0 0 0 0	
7	13	87157155	87157155156261262158155156	87 0 0 0 0 0 0 0	
7	14157155	87158155156261262158155156	87155156	87 158154 0 0 0 0 0 0	
7	10156257258157	155156	20 85156 13	0 0 0 0 0 0 0 0	
7	10	85156	20156	13 85156 13	0 0 0 0 0 0 0 0
7	2	85156	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	
8	15156	87161155	87158155156227228156267268159155	0 0 0 0 0 0 0 0	
8	15156	87158155	87159155156227228156261262158155	0 0 0 0 0 0 0 0	
8	13156	87157155	87157155156267268159155156	0 0 0 0 0 0 0 0	
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9	15156	87158155	871571551562230227228156255256	0 0 0 0 0 0 0 0	
9	15157155156	87158155	87157155156227228156255256	0 0 0 0 0 0 0 0	
9	12157155156	87157154156257258157155156	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	
9	6 20 85	62 85 84 74	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	
9	1156	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0	
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11	15	87157155	87159155156227228156255256157155156	0 0 0 0 0 0 0 0	
11	13	87157154	87157155156261262158155156	87 0 0 0 0 0 0 0 0	
11	15157155	87158155156229230156263264158155156	87	0 0 0 0 0 0 0 0	

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11	1156 0 0 0 0 0 0 0 0 0 0 0 0 0 0		
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12	13 87157154 87158155156255256157155156	87 0 0 0 0 0 0 0 0	
12	15157155 87160155156229230156255256157155156	87 0 0 0 0 0 0 0 0	
12	14157155 87157155156255256157153156	87/15/155 0 0 0 0 0 0 0 0	
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13	10202200201336337203200201 17156	0 0 0 0 0 0 0 0 0 0	
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14	15202200201247248201330331202200201202200204200	0 0 0 0 0 0 0 0	
14	1520 136337203200201202200202200201330331202200	0 0 0 0 0 0 0 0	
14	11201202200201340331202199201 20156	0 0 0 0 0 0 0 0 0 0	
15	16201202199204200201245246201330331202200201202200	0 0 0 0 0 0 0 0	
15	13202200201350351205200201202200202200201	0 0 0 0 0 0 0 0 0 0	
15	17336337203200201202200203200201245246201330331202199	0 0 0 0 0 0 0 0	
15	11201202200201334335202200201 20156	0 0 0 0 0 0 0 0 0 0	
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16	15201142343204200201202200205200201245246245246	0 0 0 0 0 0 0 0 0 0	
16	16201136337203200201202200204199201245246201336337	0 0 0 0 0 0 0 0	
16	14201200201202200201245246201336337203200201	0 0 0 0 0 0 0 0 0 0	
16	2 13156 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		
17	15201202200207200201330331202200201202200202199	0 0 0 0 0 0 0 0	
17	19201330331202200201202200202200201342343	0 0 0 0 0 0 0 0	
17	16204200201202200206199201245246201336337203200201	0 0 0 0 0 0 0 0	
17	11204200201330331202200201 20 62 85	0 0 0 0 0 0 0 0 0 0	
17	3 84 73156 0 0 0 0 0 0 0 0 0 0 0 0 0 0		
18	15201206200202200201342343204200201202200204200	0 0 0 0 0 0 0 0	
18	14201245246201330331202200201203200203200201	0 0 0 0 0 0 0 0	
18	15330331202200201202200203200201348349205200201	0 0 0 0 0 0 0 0	

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AD-A193 828

F-14 WING OUTER PANEL FATIGUE TEST SPECTRUM(U) NAVAL  
AIR DEVELOPMENT CENTER WARMINGSTER PA AIR VEHICLE AND  
CREE SYSTEMS TECHNOLOGY DIRECTORATE G S SEIDEL ET AL.  
APR 87 NADC-87056-60

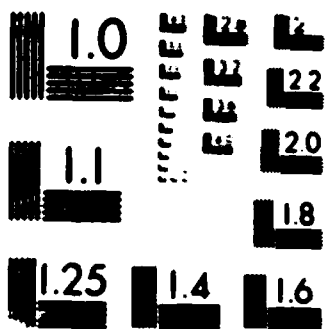
3/3

UNCLASSIFIED

F/G 1/3.3

NL





MICROCOPY RESOLUTION TEST CHART  
10-11-1963

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310	15156	87157	154	87157	155156	229230	156261	262158	155	0	0	0	0	0	0	0	0
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310	15	87159	155	17158	155156	227228	156255	256157	155156	u	0	0	0	0	0	0	0
310	15	87158	155	87157	155156	229230	156261	262158	155156	0	0	0	0	0	0	0	0
310	13	87157	155156	227228	156261	262158	155156	9	0	u	u	u	u	0	0	0	0
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