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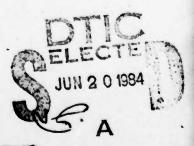


FINAL REPORT ON

DEVELOPMENT OF A DISTRIBUTED BREACH FOR THE CONICAL SHOCK TUBE

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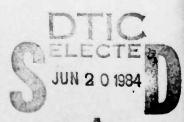
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COLLEGE OF ENGINEERING
ORLANDO, FLORIDA
January, 1983

Final Report
on
Development of a Distributed Breach
for the Conical Shock Tube

Submitted by Sayed M. Metwalli Faissal A. Moslehy

In Accordance with the Requirements of
Contract N00014-82-K-2049
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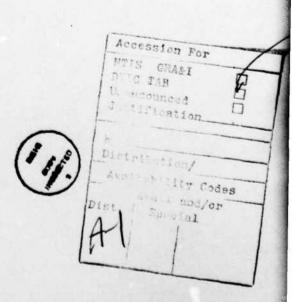
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ABSTRACT

This report represents the final stage of distributed breach development for the conical shock tube. An initial design of the distributed breach has been used to find the effect of prestressing before firing on the stress state after firing. Finite element method has been used to evaluate in-plane and hoop stresses before and after firing. A coarse finite element model is used to find points of high stresses before a finer mesh thereat is adopted. Results confirm the existance of a prestress three dimensional continuum which creates a very high resistance to firing loads. In fact, stresses have literally been improved after firing due to prestressing effect. The results of the initial design led to modifications which can further improve the stress distribution in the breach. The improved design with its working drawing is included in the redesign section of this report. The next stage is the manufacturing and testing of the improved design. This will be included in this final report of the project.

Test results indicate a marked improvement over the old tube. No failure has occured and the efficiency of simulating real blasts is about 90% which is much higher than the old tube.



ACKNOWLEDGEMENTS

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TABLE OF CONTENTS

I.	INTROD	UCTION	Page 1
II.	DESIGN	CALCULATION	3
	11.1.	Description of Initial Design	3
	11.2.	Boundary Conditions	5
		.1. Stiffness Calculation	5
		.2. Surface Pressure Before Firing	7
		.3. Surface Pressure After Firing	10
	11.3.	Finite Element Implementation	11
		.1. Coarse Finite Element Model	11
		.2. Refined Finite Element Model	11
		A. Stresses Before Firing	17
		B. Stresses After Firing	17
III.	DISTRI	BUTED BREACH REDESIGN	36
IV.	TESTING OF THE NEW BREACH		41
	IV.1	Charge Calculations	41
	IV.2	Test Results	43
٧.	CONCLUSION		50
APPENDIX A			51
APPENDIX B			65
REFERENCES			96

INTRODUCTION

A program of development and redesign of the existing shock tube is presented. One of the main problems which has limited the continued successful use of the tube is the plastic deformation in the breach resulting from the detonation. The objectives of this proposed continuation of the current program is to develop a new design to the shock tube with a distributed breach which should enhance shock wave characteristics by minimizing the energy losses associated with the plastic deformation.

An explosive driven hydrodynamic conical shock tube was developed [1, 2, 3] to test the integrity of a device in an explosive underwater environment. The original design utilized an expendable mild steel breach to confine the explosive. The number of shots which could be made before replacing the breach plug varied from a large number when using a blasting cap only to one or two when using 10 grams equivalent TNT. Also the loss of energy through the resulting plastic deformation severely limits the amplification factor - reducing it from 7770 theoretically to approximately 1400 when using 10 grams equivalent TNT. The shock tube was fabricated in two pieces of approximately four feet and six feet in length. This was done to facilitate handling of the tube, whose total weight is approximately 1200 pounds.

In this interim report the four foot section of the tube is discarded and the equivalent amount of charge is distributed over a spherical surface at that station. Current results of dynamic history response of the shorter shock tube [3] has indicated no degradation in natural frequencies or mode shapes. The dynamic stress wave in the tube wall indicated tolerable magnitudes. Stress analysis is herein initiated to investigate the proposed design.

To get a closer insight into the stresses in the distributed breach, a simple and approximate model is selected for the expendable part. After the analysis is completed accordingly a redesign of the breach is consumated depending on the findings of the preliminary analysis.

II. DESIGN CALCULATION

II-1. Description of Initial Design

The initial design is illustrated in Figure 1. Again, use of an expendable section is recommended to prevent the occurrence of damage to the body of the tube. Use of this piece will also permit the insertion of an isolation layer between the breach and the tube body.

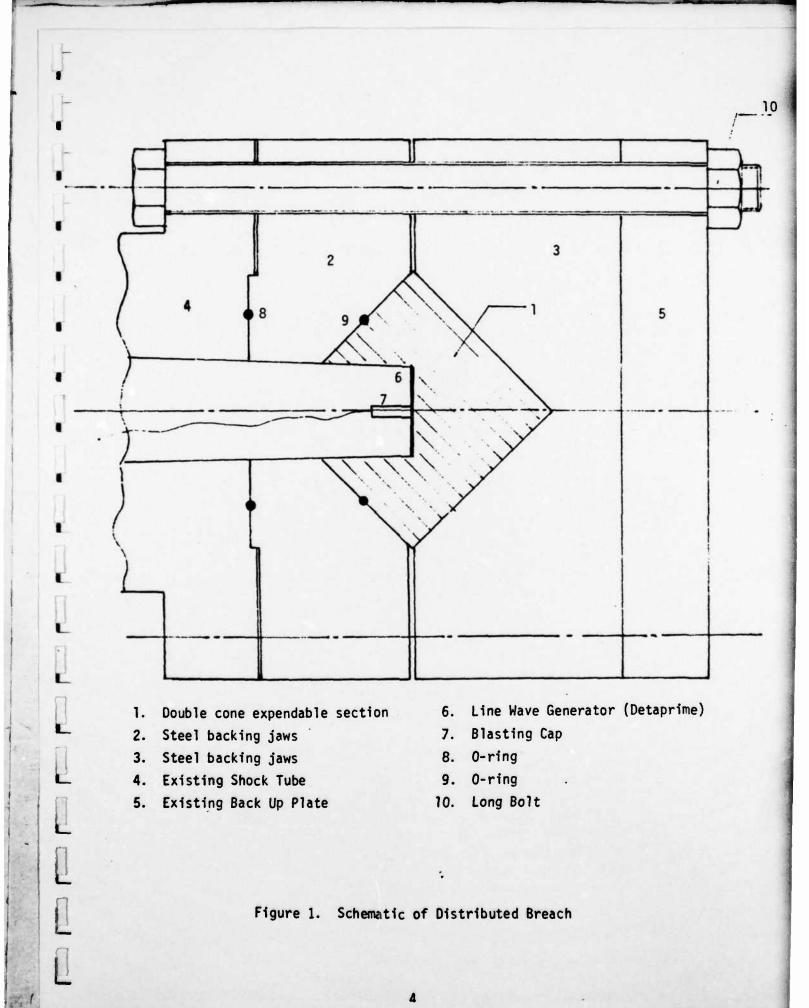
This new design provides a prestressed three dimensional continuum, part (1), Figure 1, around the detonation area which creates a triaxial state of hydrostatic stress [4]. The proposed design has, therefore, considerable advantages over the existing design.

A spherical wave front of 4' radius should exist immediately after detonation. Initiation will be by use of a single blasting cap located at the tube centerline. The rate of detonation will be approximately 20,000 fps.

The rate of propagation of the shock wave in water is approximately 5000 fps.

Therefore the surface of the distributed charge should not be spherical, but should actually be concave by almost 1/4" at the tube centerline (5000/20000). It is recommended that the initial feasibility be determined, using a flat surface. This would allow use of DuPont Line Wave Generator which is a perforated flexible explosive prepared from Datasheet. Its thickness of 0.050" is almost exactly what is needed to provide the desired total amount of explosive.

The available finite element computer program (SAP IV) [5] is used in the stress analysis of the expendable part (part 1, fig. 1). To perform the finite element analysis it is necessary to identify the boundary conditions and to define type, form, and number of element.



II-2. Boundary Conditions

Stiffness calculation of the expendable part is required to properly evaluate the external pressure before and after firing. The prestressing before firing will be altered after firing. The following calculations are performed to define both conditions.

II-2. 1. Stiffness Calculation

Figure 2(a) and (b) show diagramatic sketches of the two ends of the expendable part. The contraction $d\delta$ of an element of length dx (Fig. 2(a)) is given by

$$d\delta = \frac{P(x)}{EA(x)} dx \tag{1}$$

where E is Young's modulus and A(x) is the area of the element dx at distance x and P(x) is the total force on the element such that

$$P(x) = \frac{P_2}{\sqrt{2}} \pi (r + a) \sqrt{(r+a)^2 + x^2} = p_2 \pi (x^2 - a^2)$$
 (2)

where p is the normal press on the surface and the cone angle is 45° . The total contraction is given by

$$\delta = \int_{EA(x)}^{x=h} \frac{P(x)}{EA(x)} dx = \frac{1}{E} \int_{\pi(x^2 - a^2)}^{x=h} \frac{P_2\pi(x^2 - a^2)}{\pi(x^2 - a^2)} dx$$
x=a

$$= \frac{P_2}{E} (h-a) \tag{3}$$

The resultant force is given by

$$F = p_2(h^2 - a^2)\pi$$
 (4)

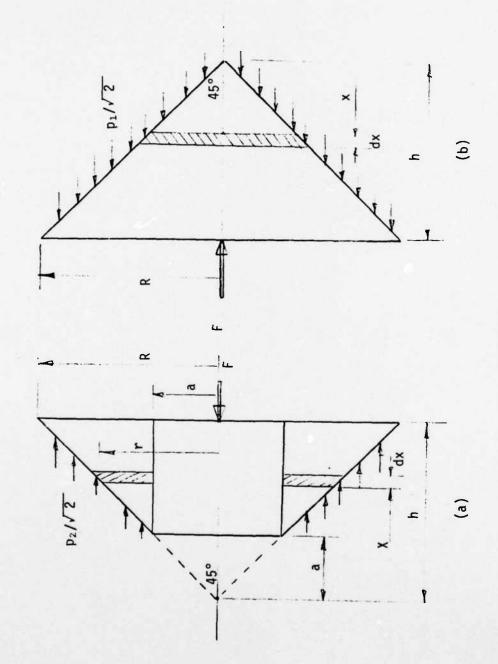


Figure 2. Diagramatic Sketch of the Two Ends of the Expendable Part

Hence the stiffness of the part shown in fig. 2(a) is

$$k_1 = \frac{F}{\delta} = \pi E(h+a) \tag{5}$$

Similarly the stiffness of the end Fig. 2(b) can be obtained by substituting a = 0 in eq. (5) to get

$$k_2 = \pi E h \tag{6}$$

The equivalent stiffness of the expendable part is then

$$k_e = \frac{k_1 k_2}{k_1 + k_2} = \frac{\pi E h(h+a)}{(2h+a)}$$
 (7)

For the initial design h is selected as 3 in. and a as 1 in. Hence eq. (7) yields

$$k_p = 162 \times 10^6 \text{ lb/in}$$
 (8)

where $E = 30 \times 10^6$ psi for steel.

L

The stiffness of a single bolt (part 10 in fig. 1) is expressed by

$$k = \frac{EA}{\ell} = 1.06 \times 10^6 \text{ lb/in}$$

where ℓ = 12.5 in and A = 0.44 in² for 3/4 in diameter bolt. The total stiffness of the 12 bolts is

$$k_b = 12 k = 12.7 \times 10^6 lb/in$$
 (9)

II. 2.2 Surface Pressure Before Firing

A diagramatic sketch of the expendable part with the surface pressure before firing is depicted in fig. 3. To evaluate the surface pressures p_1

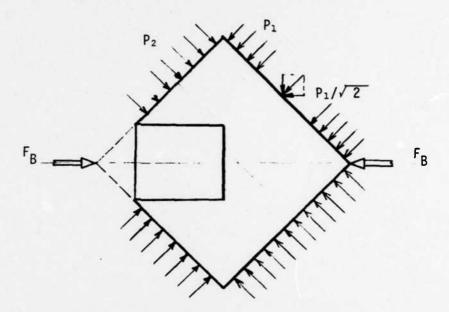


Figure 3. Diagramatic Sketch of the Expendable Part with Surface Pressure Before Fixing

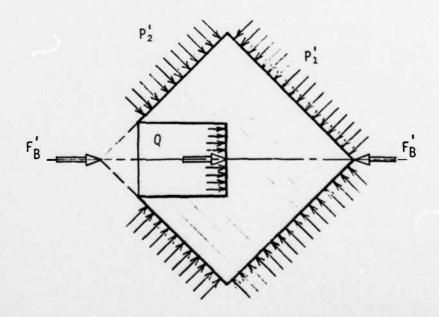


Figure 4. Diagramatic Sketch of the Expendable Part with Surface and Detonation Pressures after Firing

and p_2 the total force generated from the prestressing caused by bolt tightening should be obtained.

Using 3/4 in SAE Grade 5 bolts with proof strength Sp= 85,000 psi and ultimate strength σ_u = 120,000 psi. The initial tightening force F_i is given by [4]

$$F_i = 0.9 \text{ SpA}_t$$

where A_t is the tensile area = 0.334 in². Therefore F_i = 25,550 lb. The maximum force produced by tightening all 12 bolts is then

$$F_B = 12 F_i = 0.307 \times 10^6 \text{ lb}$$
 (10)

The normal pressure p1 generated by this force is obtained from

$$\frac{p_1}{\sqrt{2}} = \frac{F_B}{\text{surface area}}$$

where the right conical surface area is $\sqrt{2}$ πh^2 . Substituting the values of F_R and h gives

$$p_1 = 10,844 \text{ psi}$$
 (11)

From equilibrium of the whole part (Fig. (3)) the ratio $\frac{p_2}{p_1}$ is obtained by

$$\frac{p_2}{p_1} = \frac{Area of right conical surface}{Area of left conical surface}$$

from which p_2 is calculated as

$$p_2 = 12,184 \text{ psi}$$
 (12)

II. 2.3 Surface Pressure After Firing

After firing the expendable part will be subjected to the loads as shown in Fig. 4. To evaluate the new surface pressures p_1 ' and p_2 ' after firing the preloading effect should be taken into account. The forces in the bolts after firing F_B is obtained from [4].

$$F_B' = \frac{k_b Q}{k_b + k_e} + F_B$$
 (13)

where Q is the total force induced by the pressure wave and is given by

$$Q = p_{m} \cdot \pi a^{2} \tag{14}$$

 p_m is the peak pressure at the location of the distributed breach (at 4 feet from the apex of tube) and equals 25,000 psi [1]. Substituting the values from eqs. (8,9,10) and (14) into eq. (13) given

$$F_{B}' = 312 \times 10^{3} \text{ lb}$$
 (15)

The force in the expendable part is given by [4]

$$F_e = \frac{k_e Q}{k_b + k_c} - F_B$$
 (16)

which upon substitution gives

$$F_{p} = -234 \times 10^{3} \text{ lb}$$
 (17)

The normal pressure p_1 ' generated by this compressive force is obtained from

$$\frac{p_1'}{\sqrt{2}} = \frac{F_e}{\text{surface area}}$$

which gives

$$p_1' = 8,276 \text{ psi}$$
 (18)

From equilibrium of the whole part (Fig. (4)), the normal pressure on the left conical surface may be calculated by

$$p_2' = \frac{F_e - Q}{\text{left surface area}} = 6,175 \text{ psi}$$
 (19)

II.3. Finite Element Implementation

A preliminary coarse finite element model of the expendable part is used first to find the points of high stresses where a finer mesh should be used. A refined finite element model is then adopted.

II.3.1 Coarse Finite Element Model

Figure 5 shows the selected mesh for the coarse finite element model with both element and node numbers indicated. The mesh was generated by using computer program GRID [6]. Twenty-five quadrilateral axisymmetric elements based on an isoparametric formulation are used. The total number of nodes is thirty-five.

SAP IV computer program [5] was used to evaluate the stresses inside the expendable part. The three principle stresses were obtained at the center of each element. Figure 6 shows the in-plane compressive principle stresses and their orientation for the case of prestressing before firing. The third principle stress (hoop) is shown in Fig. 7.

Stresses generated after firing were also computed and displayed as shown in figures 8 and 9.

II.3.2 Refined Finite Element Model

It can easily be seen that high stress gradients occur around the detonation area. For this reason a finer mesh generated by the computer program GRID was adopted around that area. A general computer program was developed

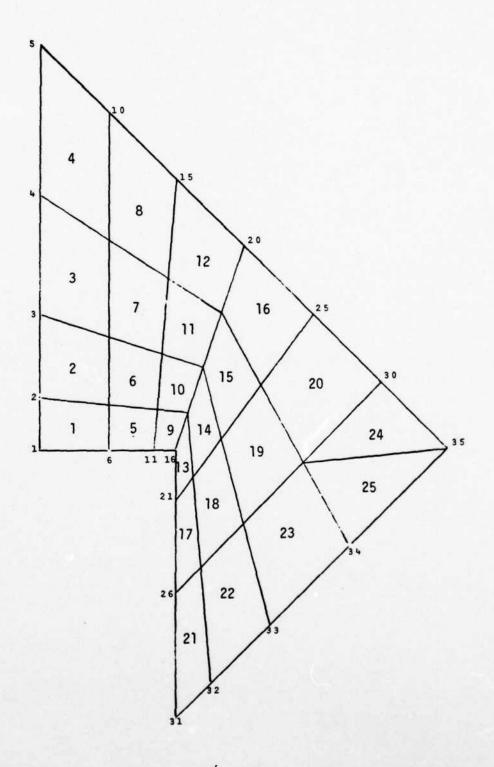


Figure 5. Coarse Finite Element Model

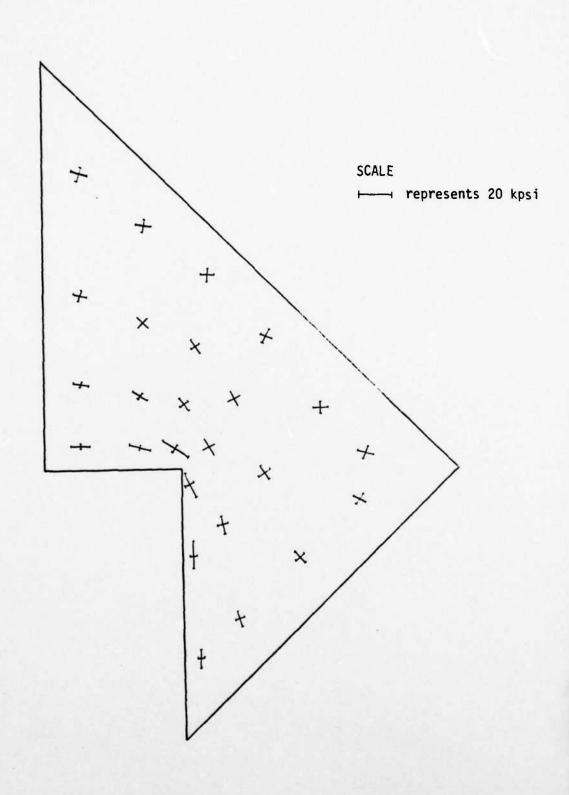


Figure 6. In-plane Compressive Principal Stresses Before Firing

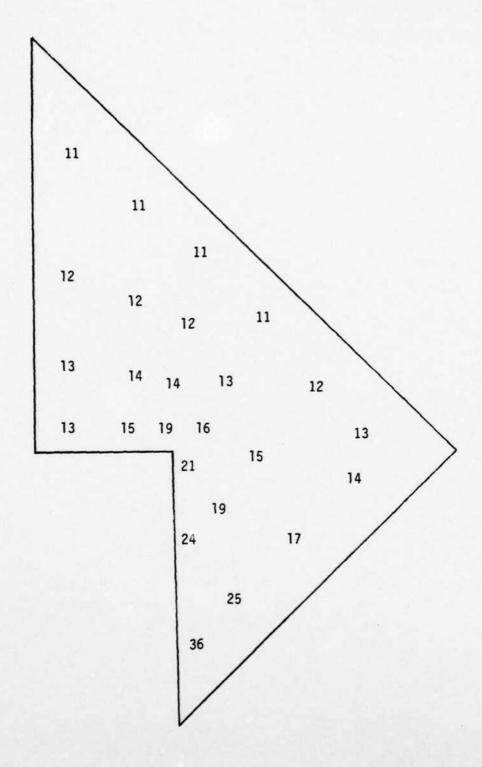


Figure 7. Third Compressive Principal (HOOP) Stresses
Before Firing

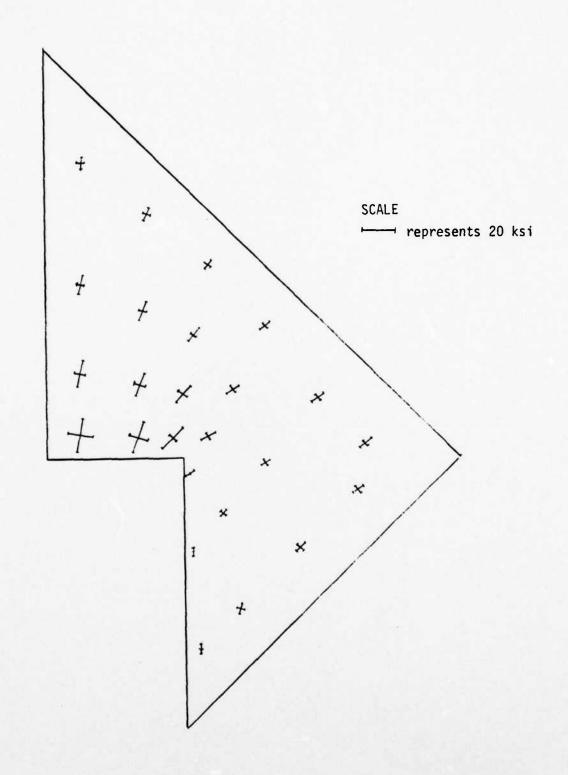


Figure 8. In-Plane Compressive Stresses
After Firing

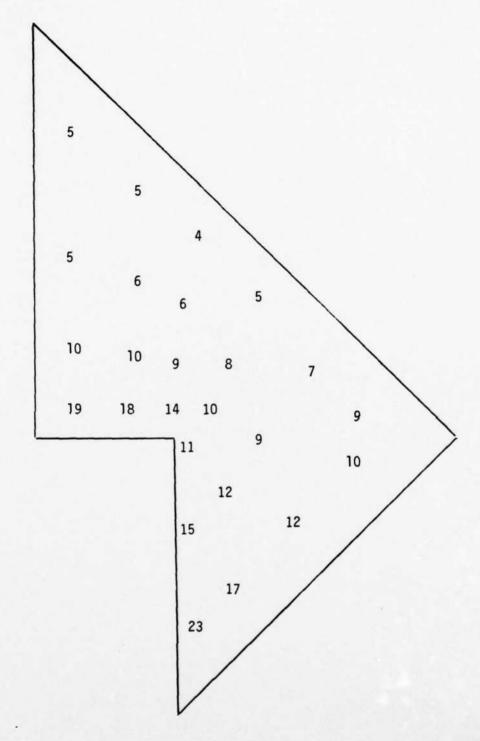


Figure 9. Third Compressive Principal (HOOP)
Stresses After Firing

to display the mesh, element number, and stress values at the center of each element. A listing of the program is included in Appendix A. Figure 10 shows the refined mesh with element numbers indicated and figure 11 depicts an enlargement of the crowded part of the mesh.

A. Stresses before firing

The output of the computer program for the three principle stresses (in-plane σ_1 and σ_2 and hoop σ_3) is plotted in figures 12-17 for the prestressing before firing. It should be noted that the printed positive values of stresses are compressive and vise versa.

Since the distortion energy theory of failure [4] is used in this analysis the value of the equivalent von Mises stress is calculated from

$$\sigma_{e} = \sqrt{\frac{1}{2}[(\sigma_{1} - \sigma_{2})^{2} + (\sigma_{2} - \sigma_{3})^{2} + (\sigma_{3} - \sigma_{1})^{2}]}$$
 (20)

Figures 18 and 19 display the values of σ_e at the center of each element. It is obvious from fig. 18 that the value of σ_e is nearly zero at the outer boundary of the expendable part except at the mouth where there is no continuity of the material. This demonstrates the state of almost hydrostatic stress and the suitability of the design of this part.

B. Stresses after firing

Principle stresses (σ_1 , σ_2 , and σ_3) computed after firing including the effect of prestressing are shown in figures 20-25. Comparing values of stresses before and after firing reveals that the state of stress has literally been improved after firing due to the prestressing effect.

The von Mises stress (σ_e) is also plotted in figures 26 and 27 which indicate the improvement in the state of stress after firing.

A complete listing of computer printout results for both before and after firing cases is included in Appendix B.

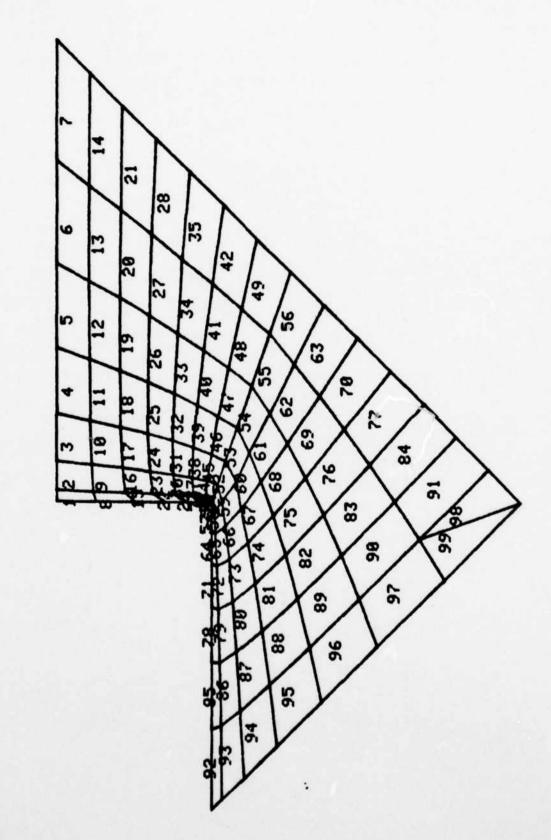
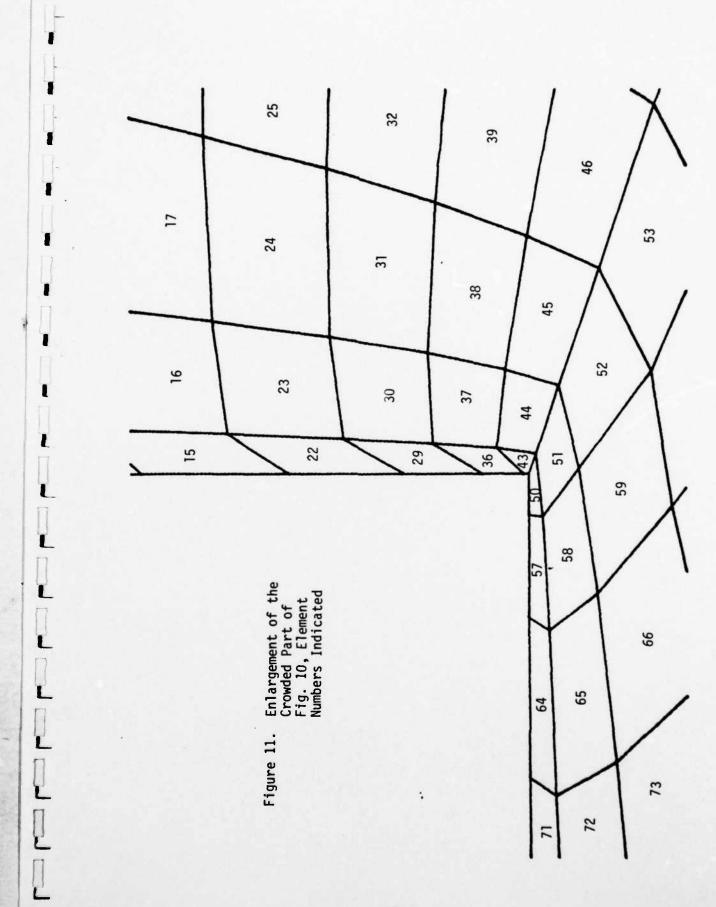
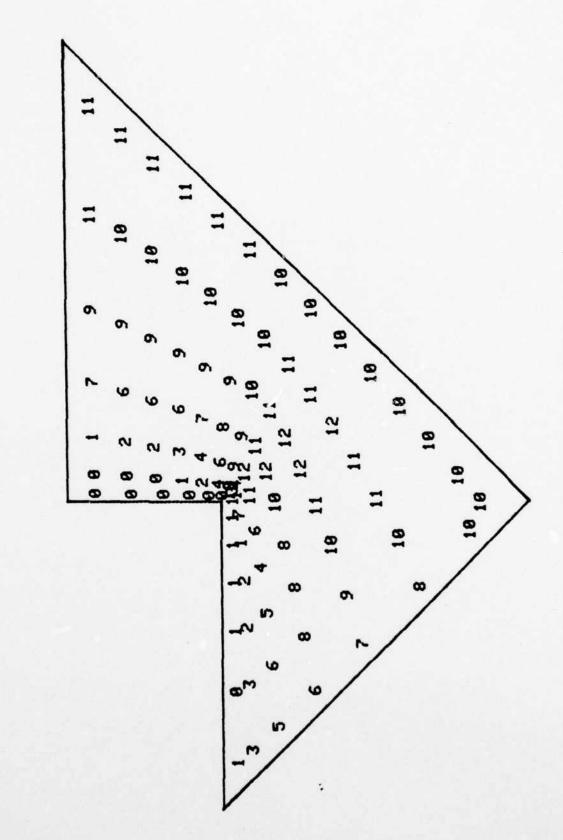


Figure 10. Refined Finite Element Mesh, Element Numbers Indicated

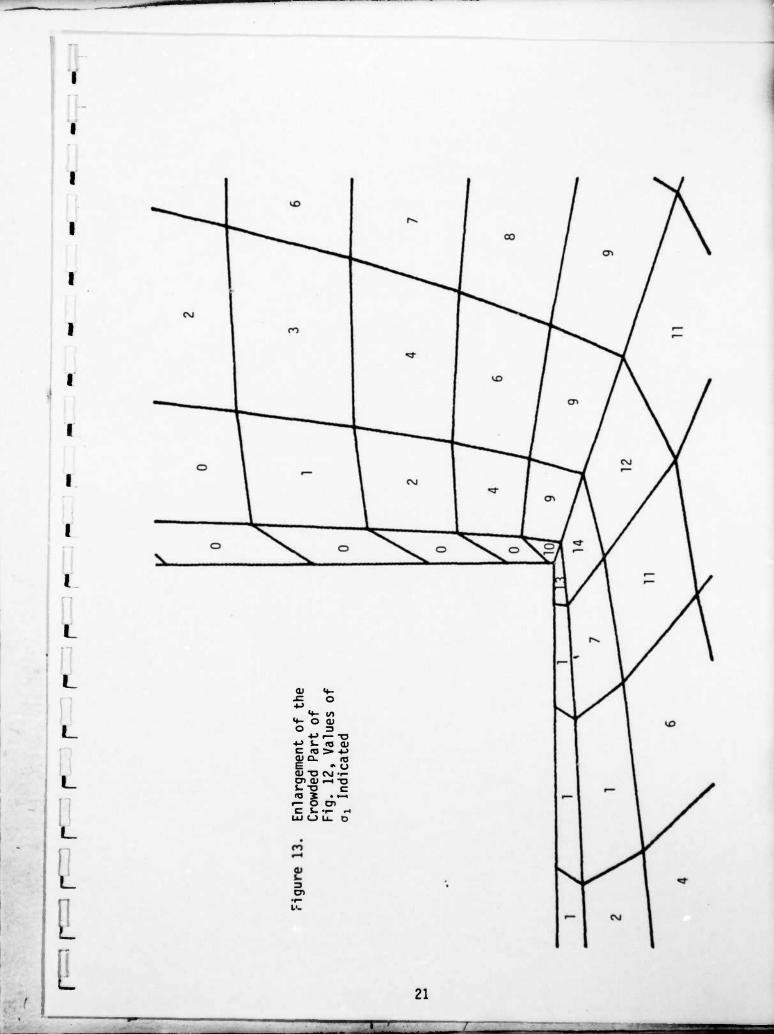




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Figure 12. Compressive Principal Stress (σ1) Before Firing



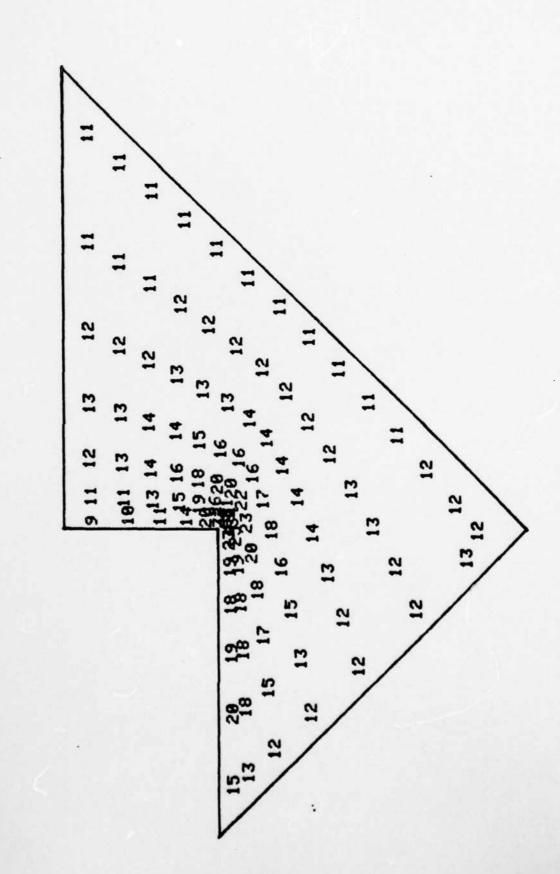
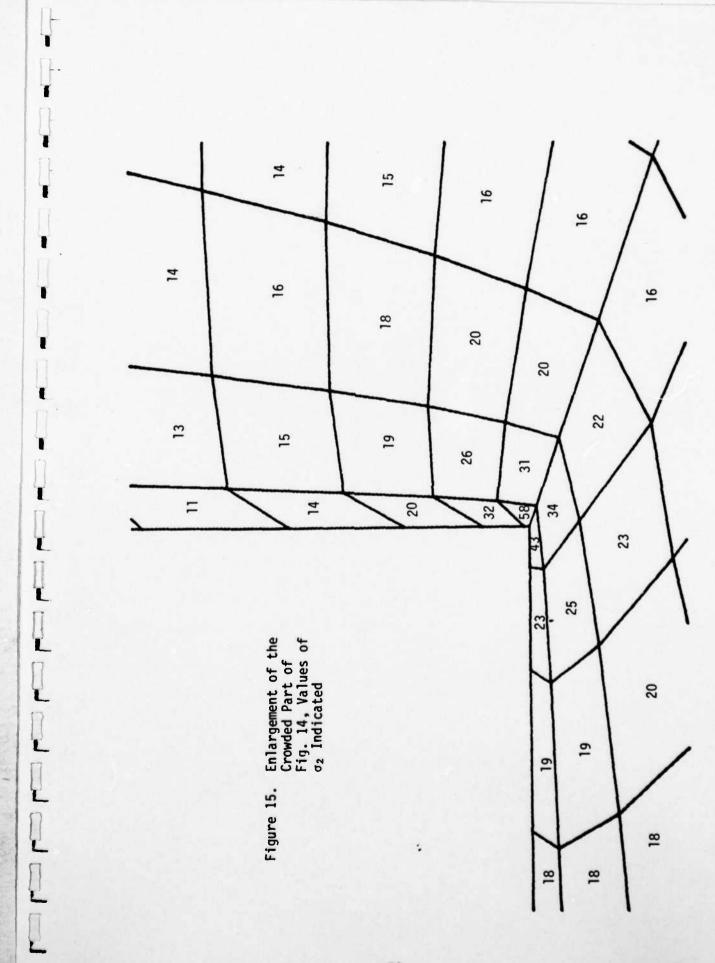


Figure 14. Compressive Principal Stress (σ₂) Before Firing



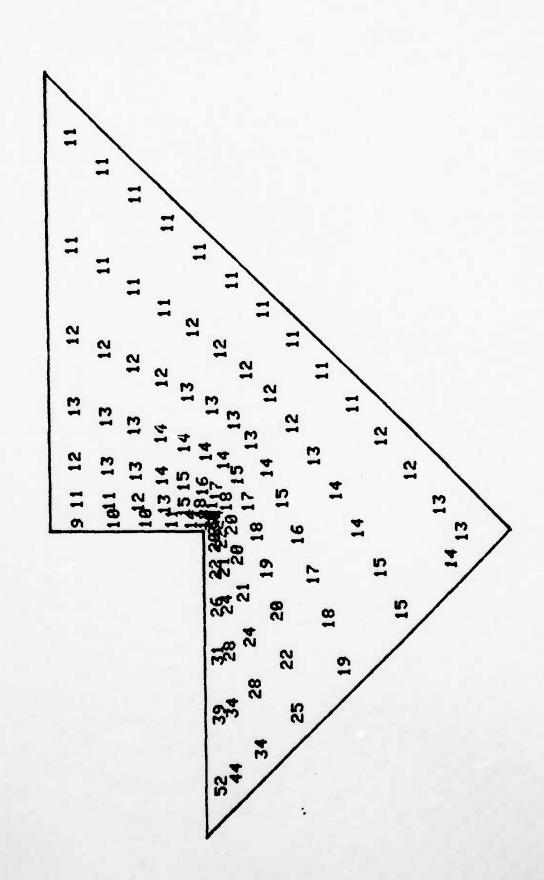
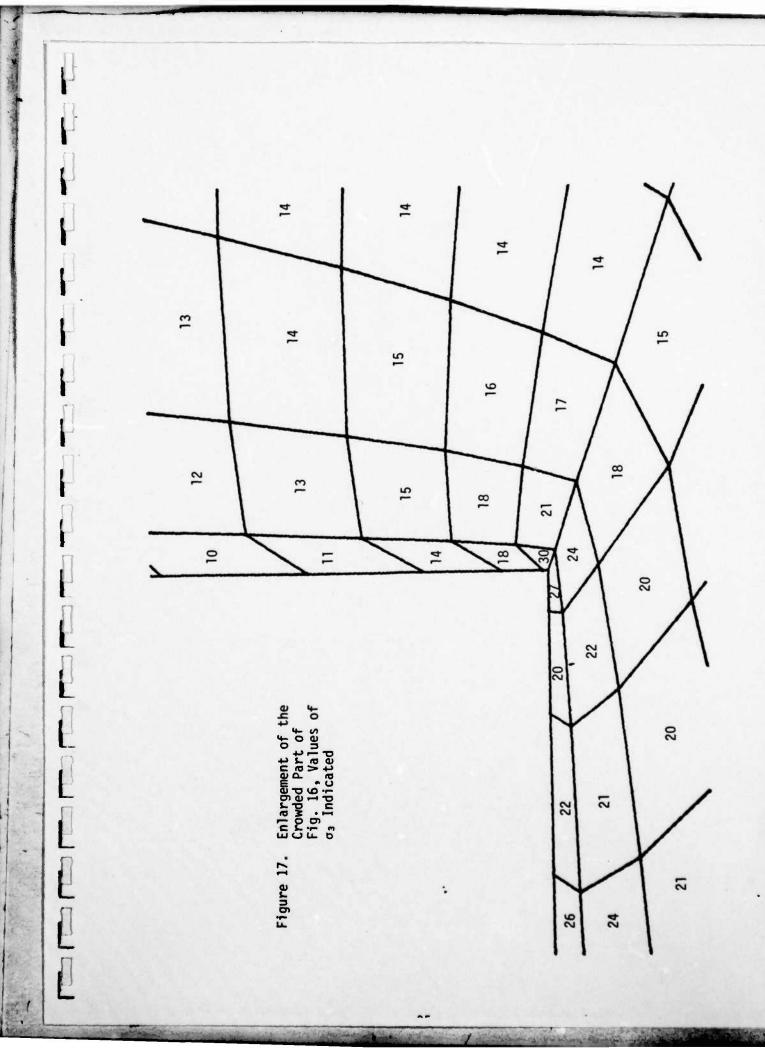


Figure 16. Compressive Principal (Hoop) Stress, 03 Before Firing



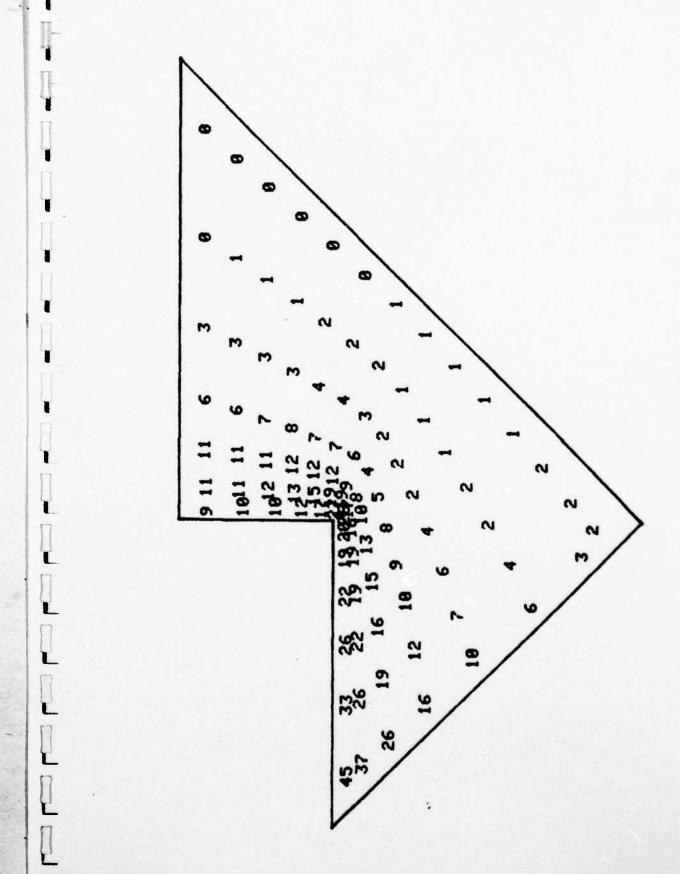
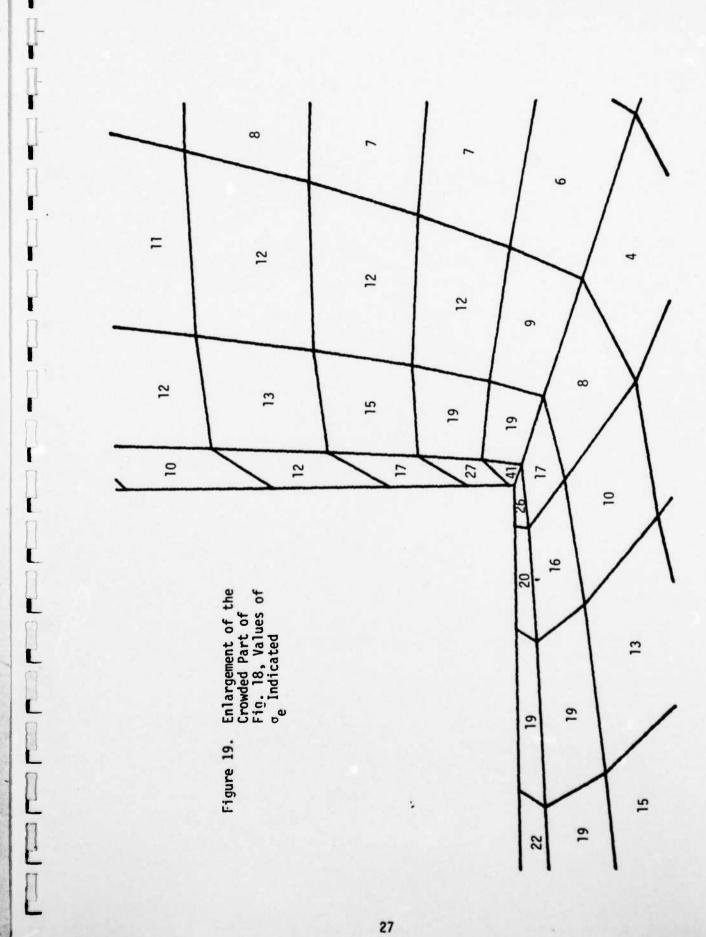


Figure 18. Compressive Equivalent von Mises Stress, $\sigma_{\rm e}$



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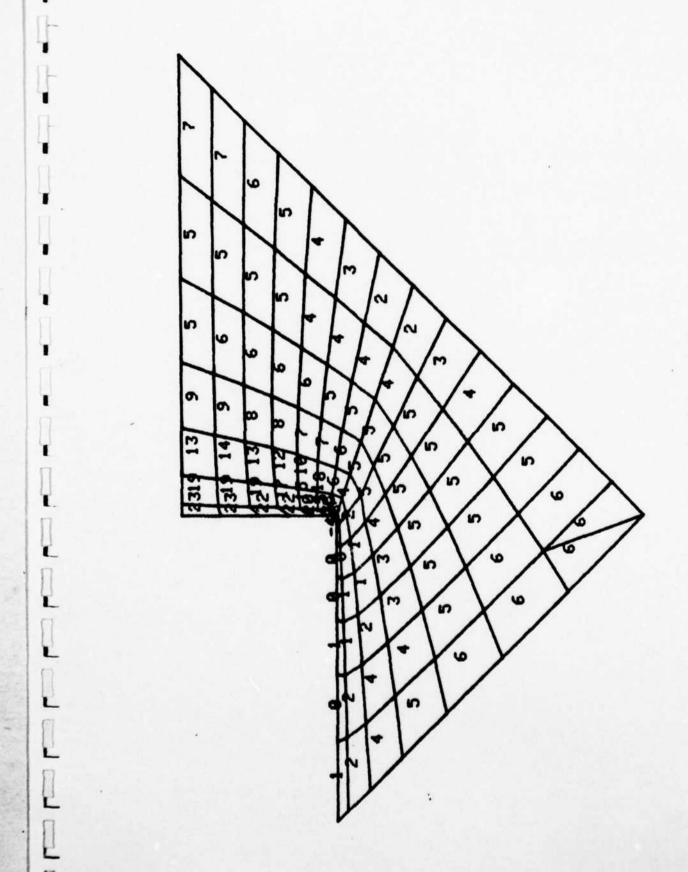
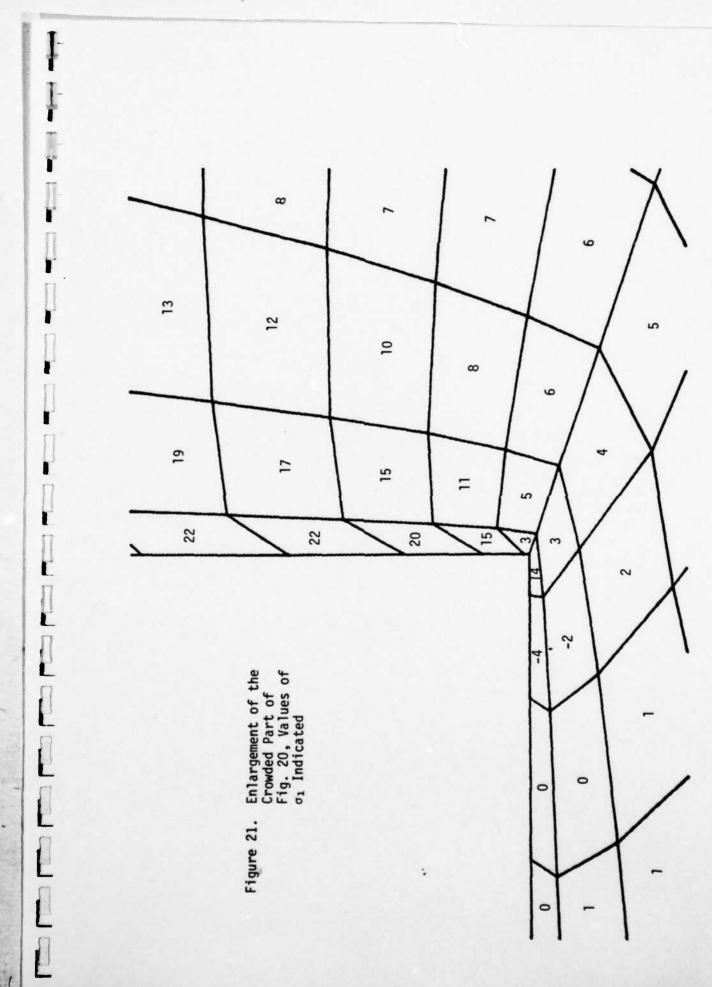
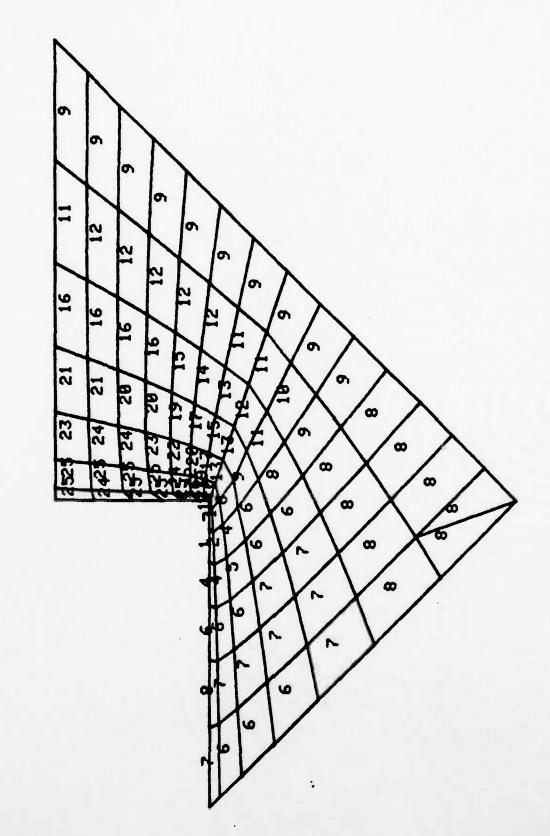


Figure 20. Compressive Principal Stress (σ_1) after Firing

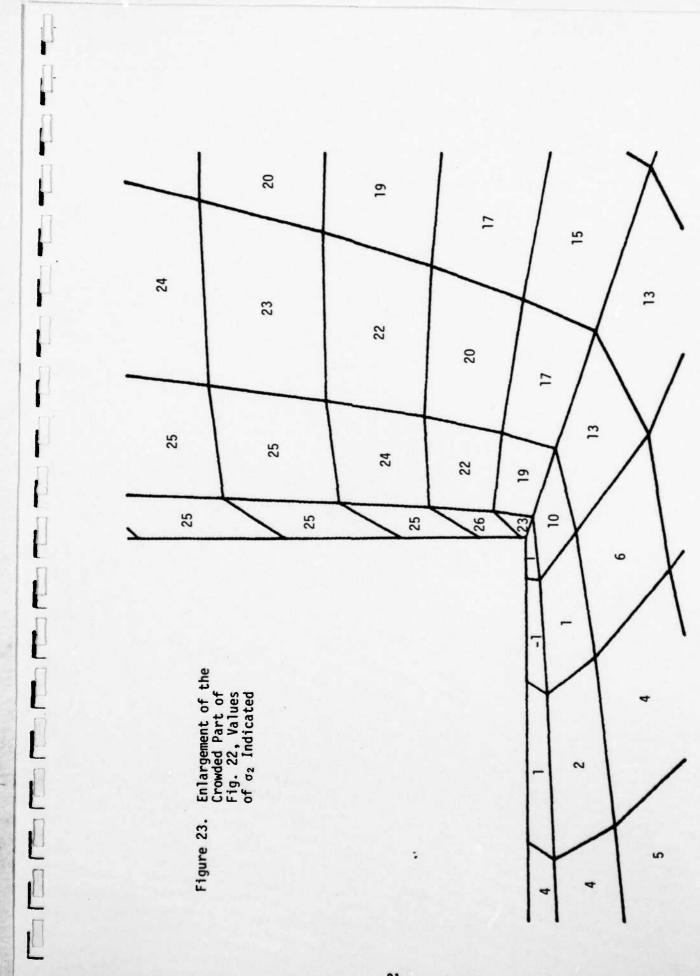




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Figure 22. Compressive Principal Stress (σ₂) After Firing



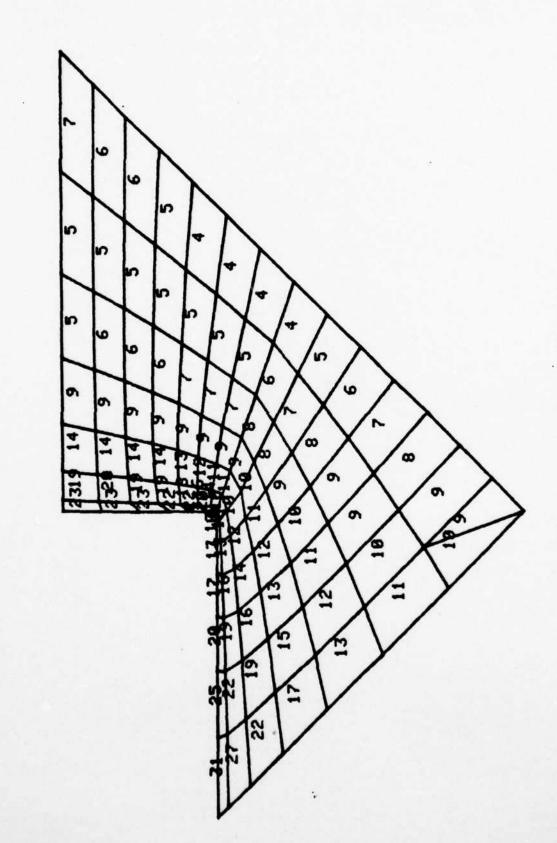
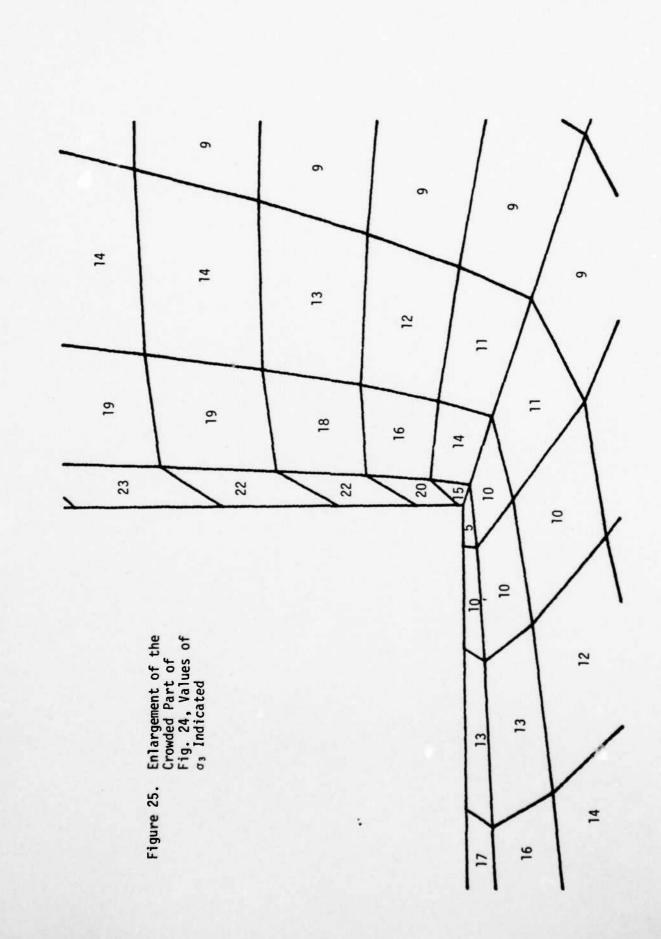


Figure 24. Compressive Principal Stress (σ_3)



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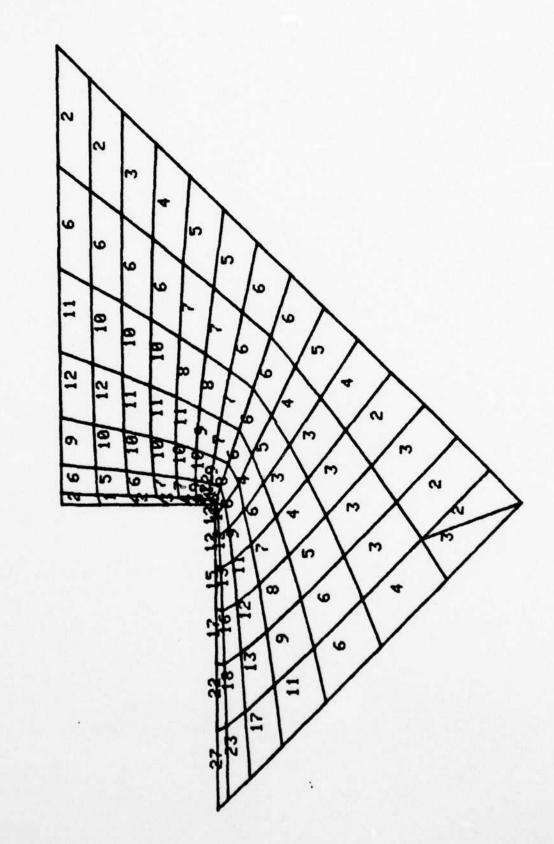
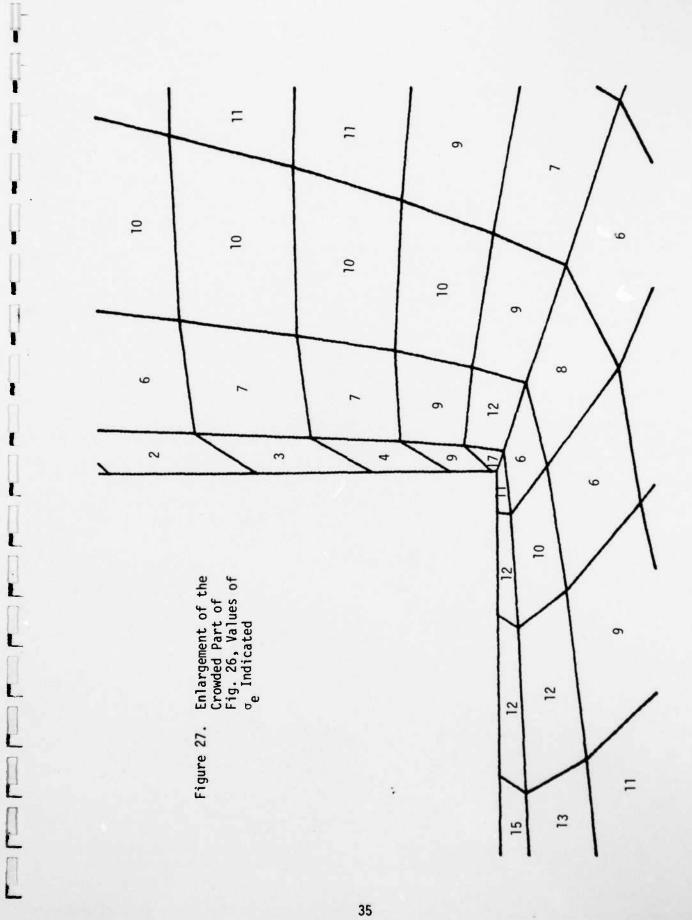


Figure 26. Compressive Equivalent von Mises Stress, σ_{e} after Firing



III. DISTRIBUTED BREACH REDESIGN

The results of the previous analysis reveal that improvement could be attained through an appropriate redesign of the breach.

Maximum utilization of material could be achieved by eliminating areas of low stresses. Furthermore, the high stress generated at the mouth of the expendable part may be reduced by the reduction of the mouth extension. Such modifications are implimented as shown in figure 28.

Detailed working drawings of distributed breach parts are included in figures 29-31.

Hot rolled G4140 steel is selected for all the parts of the distributed breach so that it would have the sufficient strength and ductility requirements. Such a material has a yield strength of 63×10^3 psi which will provide a factor of safety of 1.4 for the worst loading condition of the initial design.

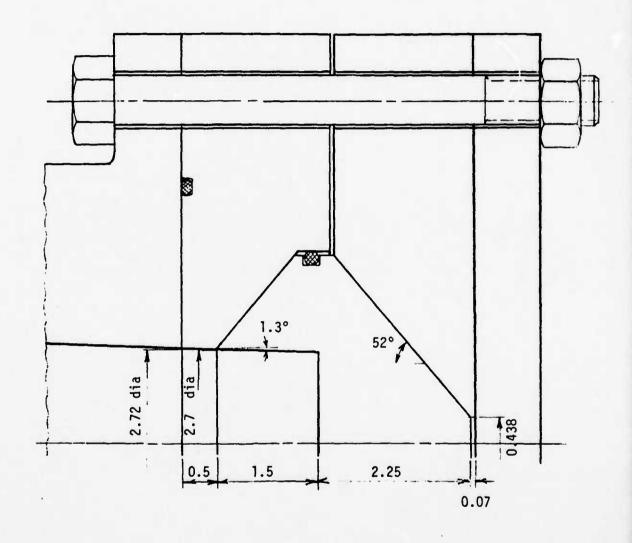


Figure 28. Redesign Modification of Distributed Breach

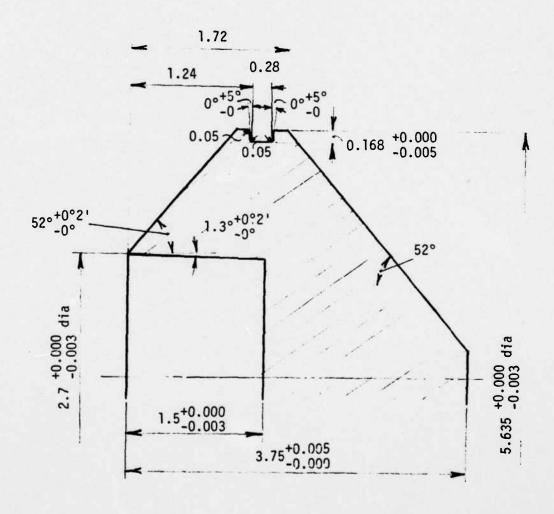
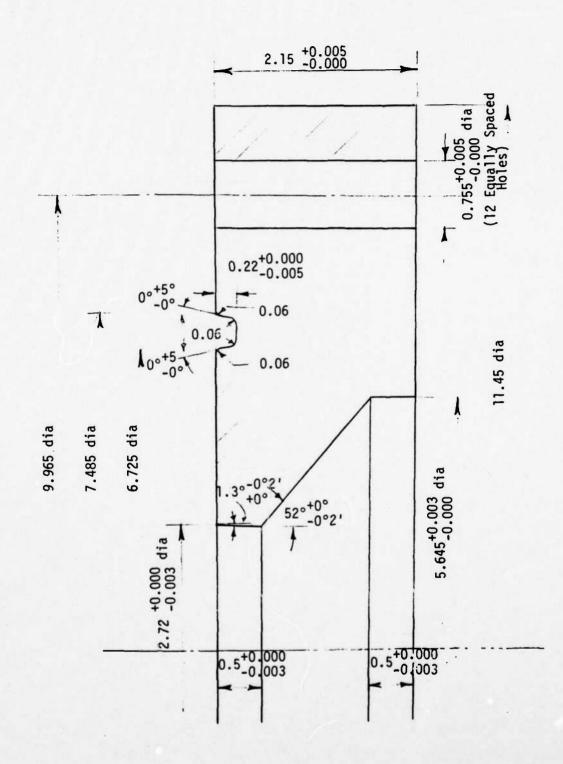


Figure 29. Detailed Working Drawing (Cross Section) of the Double Cone Expendable Section



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Figure 30. Detailed Working Drawing (Cross Section) of the Left; Backing Jaw

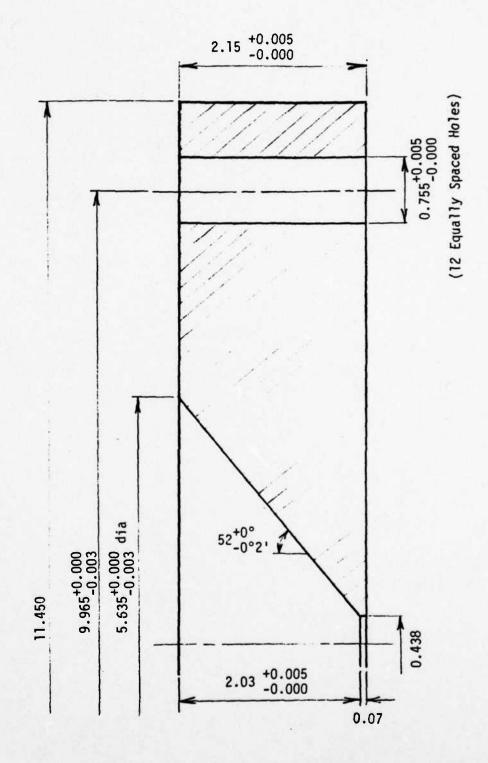


Figure 31: Detailed Working Drawing (Cross Section) of the Right Backing Jaw

IV. TESTING OF THE NEW BREACH

IV-1. Charge Calculations

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The fundamental principle behind the conical shock tube is that a small conical charge placed at the vertex of the conical tube produces the same shock effect as a spherical charge of equal radius would produce in a free field [1]. The shock tube merely isolates this portion of the wave from the rest and ideally has no effect on the wave characteristics [1]. The shock waves in the tube should be sperical in nature and the explosive energy liberated by the conical charge will be concentrated into the solid angle of the cone rather than radiating in all directions. The small amount of explosive thus behaves like a much larger amount and an amplification is realized. The amplification factor (AF) can then be defined as the weight of apparent spherical charge to that of the actual conical charge. The AF then can be viewed as the ratio of the solid angle of a sphere (4π steradians) to that of the cone.

The form of the shock wave may be approximated by a discontinuous rise in pressure followed by an exponential decay [7]

$$P(t) = P_{m} exp(t/\theta).$$
 (21)

The scaling laws are empirical correlations relating the peak pressure (P_m) , time constant (0), and other shock wave parameters to the charge weight (W) and the range (R) from the charge center. For TNT the following relations apply [8]:

$$P_{\rm m} = 2.16 \times 10^4 (W^{\frac{1}{3}}/R)^{1.13}$$
 (22)

$$\theta = 58 \text{ W}^{\frac{1}{3}} (\text{W}^{\frac{1}{3}}/\text{R})^{-0.22} \tag{23}$$

where W is the charge weight in pounds, R is the distance from the charge in feet, P_{m} is the pressure in psi, and θ is the time constant in microseconds.

The tube is to generate a shock wave whose characteristics are equivalent to that of a 125 pound spherical charge of TNT up to a range of 11.0 feet. The

scaling laws show that to match peak pressure the same value of reduced distance ($W^{\frac{1}{3}}/R$) is required. Obviously, this places no constraint on the length of the tube. However, to match the time constant (0), the same apparent weight must be used since

$$\Theta = W^{\frac{1}{3}}f(W^{\frac{1}{3}}/r). \tag{24}$$

Therefore, to match both the peak pressure and the time constant an apparent weight of 125 pounds must be used $(W_{APPARENT} = AF(W)_{TRUE})$ and the formal tube length must be at least 11.0 feet from the center of the apparent charge to the muzzle end.

From the old tube, the cone angle

$$\tan \alpha/2 = (3)in/(11)(12)in,$$
 (25)

or $\alpha = 2 \arctan (0.0227) = 2.6^{\circ}$.

The theoretical AF can now be determined from [1]

$$AF = \sin^{-2}(2.6/4) = 7770.$$
 (26)

Assuming for now that this level of amplification is achievable the true weight of explosive required is [1]

The Dupont Company manufactures a flexible sheet explosive with trade name Detasheet which is available in perforated 0.05 inch of what they denote as line wave generator. Blasting caps with a two grain strength were used to initiate the Detasheet. Since Detasheet is made of PENT, and since 1 gram of TNT is equivalent to 9.45 grains and 1 gram of PENT is equivalent to 15.4 grain, then the actual weight of the Detasheet (used for the 7.3gm TNT) should then be

$$W_{ACTUAL} = 7.3(\frac{9.45}{15.4}) = 4.48 \text{ gm}$$
 (28)

The two grains blasting cap should replace

WDETASHEET = 2 grains
=
$$\frac{2}{15.4}$$
 = 0.13gm (29)

The net required Detasheet that should generate the same energy as required at the vertex of the cone is then

$$W_{DFTASHFFT} = 4.48-0.13 = 4.35 gm$$
 (30)

With the known density of the perforated Detasheet, this weight should come from more than one layer of the explosives.

IV-2 Test Results

Five tests have been conducted at NRL-Orlando with the new design withstanding all of them without any sign of failure to the distributed breach. The only observation is the minor imprint on the surface due to the perforated pattern of the line wave generator (Fig. 32). No change in shape or dimensions has resulted and no plastic deformation occurred.

The first shot (#1) was conducted with only 2.6 gm of Detasheet. This represented only one layer of the explosives and was selected as primary test. The resulting pressure wave is shown in Fig. 33. With approximate extrapolation of the curve and using the approximate calibration of this gage (76B,-253.1 dB/1V/ μ Pa), the peak pressure was found to be approximately 7500 psi. Using the scaling laws ratio, the pressure for a full charge should have been approximately 9000 psi. The expected pressure at this distance (10 ft) for a 125 pound spherical charge of TNT is 9869 psi (see Eqn. 22). This would give an efficiency of about 91%.

The next two shots (#2 & #3) were carried out with the full charge of 4.35 gm of Detasheet. Due to probe failure in shot #2, the results were discarded. Shot #3 resulted in the pressure wave shown in Fig. 34. Performing the approximate extrapolation and calculations, we can find the pressure to be 8800 psi. This would represent an efficiency of about 89%.



Fig. 32

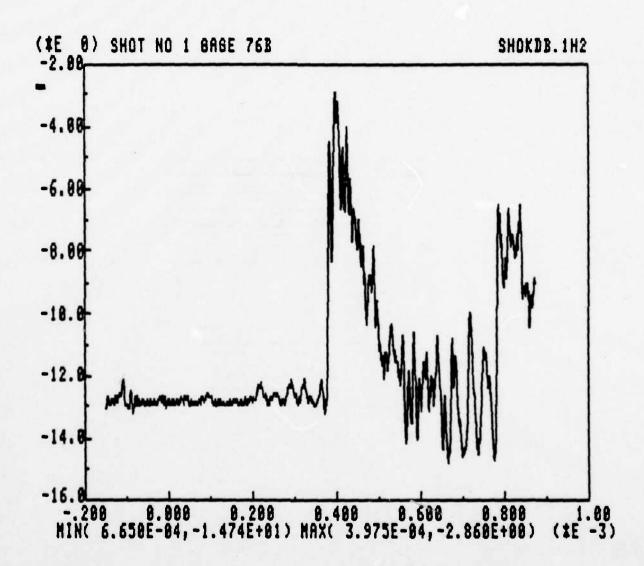


Fig. 33

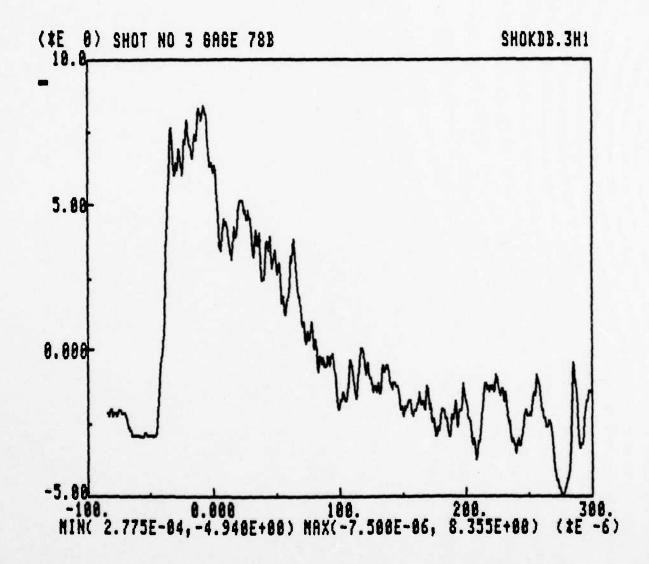


Fig. 34

The following two shots (#4 and #5) were performed with a fortified rubber gasket inserted between the breach assembly and the tube and with the full charge of 4.35gm. The results are shown in Figs. 35 and 36. Although no calibration value is available for the gage used, the pressure response has improved particularly at the peak.

In all tests, however, the pressure gage mounting has suffered from the blast and needed to be changed.

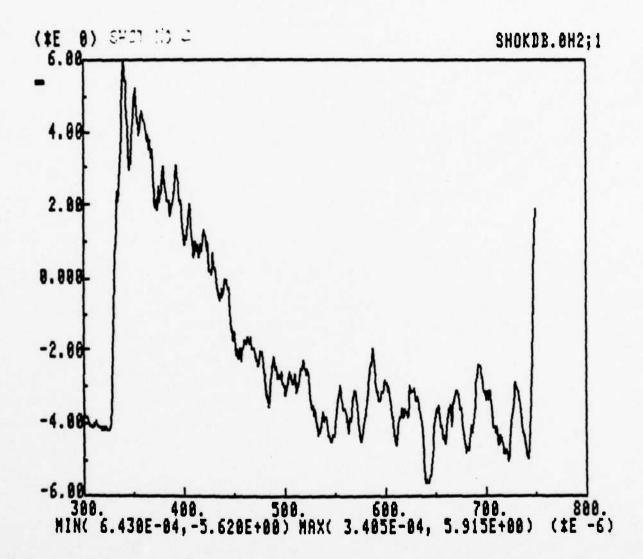


Fig. 35

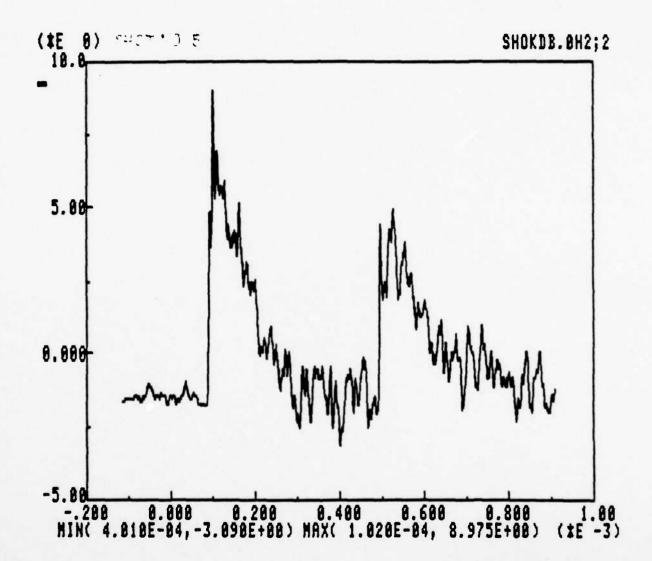


Fig. 36

V. CONCLUSION

Test results of the new distributed breach design indicate that the new design can withstand the full load without any sign of failure or damage. The efficiency of simulating real blasts is about 90% which is very high compared to the efficiency of the old tube. The pressure gages, however, have suffered from the shock wave and required replacement almost after every shot. A different type of gages might be used and a different gage mounting should be designed.

APPENDIX A

Program listing for element mesh drawing and displaying element number and stress values at element center

```
LIS

118 PAGE

118 PAGE

128 UIENPORT 8,138,98,-18

128 WINDOW 8,5,8,3.6

148 READ M,N

158 DIN X(N), Y(N), J(N,4)

168 DIN X(N), Y(N), J(N,4)

198 READ J(1,K)

228 FOR L=1 TO N

238 READ J(1,K)

248 FOR L=1 TO N

258 FOR L=1 TO N

258 FOR L=1 TO N

258 FOR I=1 TO N

268 L1=J(1,1)

338 DRAW X(N1), Y(N1)

348 HEXT K

358 II=J(1,1)

358 II=J(1,1)

368 II=J(1,1)

378 IZ=J(1,2)

388 IZ=J(1,2)
```

I

1

```
Y2=(Y(12)+Y(13))/2

X3=(X(11)+Y(12))/2

Y3=(Y(11)+Y(12))/2

X4=(X(13)+X(14))/2

Y4=(Y(13)+Y(14))/2

R1=X2-X1

R1=X2-X1

R1=X2-X1

R2=X4-X3

C2=X4X3-X3XY4

D=A1XB2-A2XB1

X5(1)=(C1XB2-C2XB1)/D

Y5(1)=(C1XB2-C2XB1)/D

Y5(1)=(C1XB2-C2XB1)/D

Y5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D

X5(1)=(C1XB2-C2XB1)/D
```

99,128 1,9,18,2,2,10,11,3,3,11,12,4,4,12,13,5,5,13,14,6 6,14,15,7,16,8,7,15,9,17,18,10,10,18,19,11,11,19,20,12 12,20,21,13,13,21,22,14,14,22,23,15,24,16,15,23,17,25,26,18 18,26,27,19,19,27,28,20,20,28,29,21,21,29,30,22,22,30,31,23 32,24,23,31,25,33,34,26,26,34,35,27,27,35,36,28,28,36,37,29 29,37,38,30,30,38,39,31,40,32,31,39,33,41,42,34,34,42,43,35 35,43,44,36,36,44,45,37,37,45,46,38,38,46,47,39,48,40,39,47 LIS
90 REM ELEMENT GRAPHIC FOR SHOCK TUBE
200 PAGE
250 UIEMPORT 0,130,90,-10
300 HOUDON 0,5,0,3,6
400 READ M,N
420 DIN X(N), Y(N), J(N,4)
500 FOR I=1 TO 4
500 READ J(I,K)
600 FOR I=1 TO 4
700 READ J(I,K)
890 NEXT K
1000 FOR L=1 TO N
1100 READ Y(L), X(L)
1200 HOUT X(NI), Y(NI)
1500 HOUE X(NI), X(NI), X(NI)
1500 HOUE X(NI), X(NI), X(NI)
1500 HOUE X(NI), X(NI), X(NI)
1500 HOUE X(NI), X(NI), X(NI), X(NI)
1500 HOUE X(NI), X(N

9, 59, 58, 59, 51, 51, 59, 69, 52 5, 64, 56, 55, 63, 57, 65, 66, 58 1, 61, 69, 78, 62, 62, 70, 71, 63 7, 67, 75, 76, 68, 68, 76, 77, 69 9, 73, 81, 82, 74, 74, 82, 83, 75 8, 78, 86, 87, 79, 88, 89, 79, 87 4, 84, 92, 93, 85, 85, 93, 94, 86 8, 90, 98, 99, 91, 91, 99, 100, 92 103, 95, 104, 96, 95, 103 2, 104, 103, 111 8, 119, 111, 110

```
2668 DATA 8.412, 3.322
2678 DATA 8.412, 3.322
2688 DATA 8.429, 4.571
2788 DATA 8.625, 2.193
2738 DATA 8.639, 2.77
2778 DATA 8.681, 3.289
2778 DATA 8.681, 3.289
2778 DATA 8.681, 3.289
2778 DATA 8.643, 4.357
2778 DATA 8.643, 4.357
2878 DATA 8.846, 2.397
2878 DATA 8.846, 2.397
2878 DATA 8.851, 2.397
2878 DATA 8.851, 2.397
2878 DATA 8.851, 2.397
2878 DATA 8.881, 2.839
2878 DATA 8.881, 2.839
2978 DATA 8.989, 2.529
2978 DATA 8.961, 2.333
2978 DATA 8.961, 2.333
2978 DATA 8.961, 2.333
2978 DATA 1.871, 2.353
2978 DATA 1.871, 2.353
2978 DATA 1.871, 2.353
2978 DATA 1.86, 2.31
```

```
3818 DATA 1.689, 2.826
3828 DATA 1.838, 2.115
3828 DATA 1.838, 2.115
3828 DATA 1.65, 2.481
3828 DATA 1.253, 2.758
3828 DATA 1.253, 2.758
3128 DATA 1.156, 2.134
3228 DATA 1.1693, 1.693
3228 DATA 1.229, 1.737
3228 DATA 1.229, 1.737
3238 DATA 1.229, 1.737
3238 DATA 1.229, 1.737
3238 DATA 1.229, 1.737
3238 DATA 1.323, 1.38
3338 DATA 1.352, 1.316
3338 DATA 1.352, 1.316
```

```
3360 DATA 1.276, 1.472
3380 DATA 1.47, 1.662
3380 DATA 1.47, 1.662
3480 DATA 2.357, 2.643
3410 DATA 1.81, 8.955
3420 DATA 1.869, 8.954
3430 DATA 1.823, 1.656
3450 DATA 1.323, 1.656
3460 DATA 1.323, 1.656
3460 DATA 1.83, 1.656
3500 DATA 1.83, 1.656
3550 DATA 1.87, 8.812
3550 DATA 1.37, 8.812
3550 DATA 1.284, 8.284
3550 DATA 1.284, 8.284
3560 DATA 1.284, 8.284
3560 DATA 1.865, 8.865
3560 DATA 1.865, 8.865
3560 DATA 1.865, 8.865
3560 DATA 1.865, 1.369
3560 DATA 1.865, 8.865
```

STRESSES BEFORE FIRING

3786 DATA 12:14:13
3726 DATA 12:12:12:13
3726 DATA 12:12:13
3736 DATA 12:12:13
3756 DATA 12:14:15
3776 DATA 12:14:15
3776 DATA 11:14:16
3776 DATA 11:14:16
3776 DATA 11:14:16
3816 DATA 11:13:14
3826 DATA 11:13:14
3826 DATA 11:13:12
3826 DATA 13:12:12
3826 DATA 13:12:12
3826 DATA 13:12:12
3926 DATA 13:12:13

3350 DATA 9,13,13
3350 DATA 10,13,13
3350 DATA 10,13,13
3420 DATA 11,11,11,11
3420 DATA 6,20,16
3450 DATA 6,20,16
3450 DATA 11,11,11
3450 DATA 10,12,12
3450 DATA 10,12,12
3550 DATA 10,12,12
3550 DATA 11,12,12

2030 DATA 23.25.23
2040 DATA 23.25.23
2040 DATA 19.25.19
2050 DATA 13.25.19
2050 DATA 13.25.19
2050 DATA 13.25.19
2050 DATA 13.23.14
2100 DATA 23.24,23
2110 DATA 19.25.20
2120 DATA 23.24,23
2110 DATA 19.25.20
2120 DATA 19.25.19
2130 DATA 19.25.19
2210 DATA 19.25.19
2220 DATA 19.25.19
2220 DATA 6.16.6
2220 DATA 12.23.14
2230 DATA 6.16.6
2230 DATA 6.16.6
2230 DATA 6.16.6
2230 DATA 6.16.6
2330 DATA 6.16.5

2388 DATA 11,22,16
2488 DATA 11,22,16
2428 DATA 11,22,16
2428 DATA 7,17,9
2458 DATA 3,9,4
2458 DATA 3,23,15
2488 DATA 3,23,15
2488 DATA 5,13,11
2588 DATA 6,15,9
2558 DATA 4,11,5
2558 DATA 4,11,5
2558 DATA 2,9,4
2558 DATA 4,11,6
2558 DATA 2,9,4
2658 DATA 2,11,8
2658 DATA 2,9,10
2658 DATA 2,9,11,13
2658 DATA 3,9,5
2658 DATA 5,11,13
2658 DATA 5,11,13
2658 DATA 5,11,13
2659 DATA 5,11,13
2659 DATA 5,11,13
2659 DATA 5,11,13

2758 DATA 1,4,16
2758 DATA 1,4,16
2778 DATA 1,5,116
2778 DATA 1,5,116
2778 DATA 1,5,116
2828 DATA 2,6,12
2868 DATA 2,6,13
2878 DATA 2,6,13
2878 DATA 2,7,13
2878 DATA 2,7,13
2978 DATA 4,7,13
2978 DATA 4,7,13
2978 DATA 4,7,13
2978 DATA 6,8,18
2988 DATA 6,8,18
2988 DATA 6,8,13
3828 DATA 6,8,13
3828 DATA 6,8,113
3828 DATA 6,8,113
3828 DATA 6,8,113

APPENDIX B

Results of SAP IV Computer Program - Before and After Firing

	ANOLE	1.00.346 1.00.346 1.00.346		4	88891 8.4.60.0. 447.98		ANOLE	89.77 89.35 88.27 1.87		ANOLE	188.94 183.83 12.05 10.02		ANOLE	83.68 86.12 80.76 15.06 18.47		ANGLE	87. 10 -86. 87 81. 02
	NIH-8	-0. 91451D 04 -0. 92564D 04 -0. 91044D 04 -0. 83883D 04 -0. 99501D 04		8-HI	-0. 10877D 05 -0. 10820D 05 -0. 10949D 05 -0. 94794D 05 -0. 12351D 05		S-HIN	-0.12036D 05 -0.1389D 05 -0.12737D 05 -0.12166D 05		S-MIN	-0.13327D 05 -0.14075D 05 -0.12075D 05 -0.15077D 05 -0.11725D 05		NIM-R	-0. 12023D -0. 12023D -0. 13843D -0. 13843D -0. 107492D -0. 107499		S-MIN	-0.11188D 05 -0.11318D 05 -0.11061D 05 -0.11805D 05
	S-MAX	0. 27631D 02 -0. 14026D 03 0. 29467D 03 0. 46015D 03 -0. 43215D 03		S-MAX	-0. 11518D 03 -0. 12522D 02 -0. 19900D 03 0. 97081D 03		S-MAX	-0. 67051D 03 -0. 91591D 03 -0. 53885D 04 -0. 65952D 03 -0. 60894D 03		S-MAX	-0. 72162D 04 -0. 86138D 04 -0. 86729D 04 -0. 61631D 04		S-MAX	-0. 9423950 -0. 9423950 -0. 9636390 -0. 1606390 -0. 845290 -0. 845290 -0. 845290 -0. 845290 -0. 945290 -0. 945		S-MAX	-0.106260 05 -0.107600 05 -0.104670 05
	S12	0. 7007BD 02 0. 64017D 03 0. 152BBD 04 -0. 69363D 02 0. 18213D 03		212	0. 29080D 03 0. 66036D 03 0. 61617D 03 -0. 763066D 02 0. 763050 03		512	0.45948D 02 0.96188D 02 0.31226D 03 0.37453D 03 0.27073D 04		212	-0. 11320D 03 -0. 39427D 03 0. 29248D 03 0. 21083D 04 0. 11372D 04		515	0.30435D 03 0.18732D 03 0.44578D 03 0.82525D 03 0.68987D 03		812	0.28406D 02 -0.30417D 02 0.91570D 02
	233	-0. 91297D 04 -0. 92112D 04 -0. 90399D 04 -0. 99180D 04		833	-0.10826b 05 -0.10779b 05 -0.10877b 05 -0.4507b 05 -0.12274b 05		833	-0.12040D 05 -0.11388D 05 -0.12747D 05 -0.12129D 05		833	-0.13420D 09 -0.12743D 09 -0.12743D 09 -0.13632D 09 -0.1464D 09		833	-0.12082D 05 -0.12187D 05 -0.11968D 05 -0.13348D 05		833	-0 112390 05 -0 113170 05 -0 11 1560 05
	252	0.27096b 02 -0.18344b 03 0.39045b 02 -0.83878b 04 -0.99467b 04		. \$25	-0. 12304D 03 -0. 233023D 02 -0. 24332D 03 -0. 122992D 04		822	-0. 67070D 03 0. 91515D 03 -0. 23980D 04 -0. 12154D 05		S22	-0. 72183D 04 -0. 86424D 04 -0. 16833D 04 -0. 148372D 05		225	-0. 92732b 04 -0. 94356b 04 -0. 91031b 04 -0. 13070b 05		525	-0.106270 05 -0.107620 05 -0.104820 05
-		-0. 91446D 04 -0. 92112D 04 -0. 88488D 04 -0. 43566D 03	8	511	-0. 10870D 03 -0. 10933D 03 -0. 6933D 03 -0. 10932D 03	6	811	-0.12036b 05 -0.11388b 05 -0.67173b 03 -0.12954b 04	7	811	-0.133250 -0.133250 -0.123670 -0.877270 -0.638720 -0.6387	3	811	-0.11989D 05 -0.12187D 05 -0.11771D 05 -0.66830D 04	(9	511	-0.11187D 05
C	LOAD LOC	אויררה הוגויא	ELEMENT (LOAD LOC	THACFO THACH	ELEMENT (LOAD LOC	ייייייייייייייייייייייייייייייייייייי	ELEMENT (LOAD LOC	PERENT PE	ELEMENT (LOAD LOC	איינרם היינדם היינדים	ELEMENT (LOAD LOC	877.

MULE	4 1 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	ANGLE	1.478 9.04.0.0 9.08.0.0 9.08.0.0 8.08.0.0	ANOLE	88.0 80.0 81.0 81.0 84.0 84.0 84.0 84.0 84.0 84.0 84.0 84	ANOLE	81.39 79.37 78.76 -3.14	ANOLE	77. 75 78. 13 75. 32 8. 33 8. 35	ANOLE 76.29 73.36 73.44 11.86	ANGLE 73. 41 20. 77 120. 73
8-14-S	-0. -0. -0. -0. -0. -0. -0. -0. -0. -0.	Z E	-0. 97073D 04 -0. 90721D 04 -0. 10467D 05 -0. 89365D 04 -0. 10642D 05	NIE B	-0.11174D 05 -0.11046D 05 -0.11326D 05 -0.99412D 04 -0.12458D 05	N. H.	-0.12970D 05 -0.12889D 05 -0.13039D 05 -0.12916D 05 -0.13026D 05	Z I I	-0.132190 -0.131500 -0.132740 -0.147030 -0.17220 -0.17220	8-HIN -0.122700 -0.121670 -0.121740 -0.137740 -0.107970	8-MIN -0.112780 03 -0.111920 03 -0.119840 03 -0.119840 03
I xy	-0.108000 05 -0.10760 05 -0.108200 05 -0.108500 05	S-M-R	-0. 39497D 03 0. 11898D 04 -0. 20701D 04 0. 24333D 03 -0. 11623D 04	X-M-S	-0. 20518D 03 0. 11137D 03 -0. 52780D 03 -0. 97280D 03	X X Y	-0.21553D 04 -0.23730D 04 -0.2374D 04 -0.2274D 04	S-H-R	-0.36736D 04 -0.38796D 04 -0.64836D 04 -0.48372D 04	S-MAX -0.90587D 04 -0.818157D 04 -0.99628D 04 -0.81387D 04	0-1-0-1-0-1-0-1-0-1-0-1-0-1-0-1-0-1-0-1
812	-0.15278D 02 -0.16510D 01 -0.27406D 02 -0.17316D 02	S	0. 17076D 03 0. 28654D 04 0. 41986D 04 -0. 36639D 03 0. 36834D 03	S. S.	0. 69343D 03 0. 17316D 04 0. 13600D 04 -0. 43326D 03 0. 36734D 03	\$12	0.16005D 04 0.19471D 04 0.20431D 04 -0.59212D 03 0.78994D 03	S12	0. 13641D 04 0. 13438D 04 0. 18135D 04 0. 33035D 01 0. 98607D 03	\$12 0. 73991D 03 0. 62107D 03 0. 68848D 03 0. 53460D 03 0. 53460D 03	\$12 0.130910 0.248180 0.348180 0.348180 0.348180 0.348180
633	-0 10831609 -0 108330 09 -0 108330 09 -0 107200 09	833	-0.95072D 04 -0.87691D 04 -0.10265D 09 -0.87455D 04 -0.10402D 05	833	-0.10936D 05 -0.11103D 05 -0.96962D 05 -0.12270D 05	833	-0.127410 05 -0.126650 05 -0.127250 05 -0.127250 05	833	-0.120300 05 -0.128620 05 -0.131450 05 -0.144430 05	\$33 -0.12221D 09 -0.12162D 09 -0.13534D 09 -0.13634D 09	S33 -0.11298b 05 -0.11275b 05 -0.11975b 05 -0.11975b 05
852	100.10063380 05000000000000000000000000000000000		-0.398100 03 0.315190 03 -0.628990 04 -0.894180 04 -0.106270 05	855	-0.249450 -0.164180 03 -0.758080 03 -0.92140 04 -0.124300 05	822	-0.231530 04 -0.231530 04 -0.128830 05 -0.129680 05	822	-0. 60132D 04 -0. 58040D 04 -0. 63233D 04 -0. 14703D 05	522 -0.92391D 04 -0.91493D 04 -0.13671D 05 -0.10684D 05	\$22 -0 1059460 05 -0 104650 05 -0 118320 05
L IS	-0.1080770 -0.108360 -0.10	S	00000	93	-0.111290 -0.110960 0.533170 -0.120960 0.533170 0.533170 0.533170 0.533170 0.533170 0.533170 0.533170 0.533170 0.533170 0.533170	10)	-0.127270 05 -0.125310 05 -0.212650 04 -0.227460 04	11)	-0. 128790 05 -0. 128260 05 -0. 128000 05 -0. 648360 04 -0. 900190 04	S11 -0.120900 -0.122410 -0.108800 -0.100690 -0.825090 05	811 -0.112280 09 -0.113260 09 -0.110650 09 -0.110650 09
רטאם רסכ	22224 7477 7477	ELEMENT (744CF0	ELEMENT (ELEMENT (הייייי הייייר הייייי	ELEMENT (ברברה החירה הוגוח	ELEMENT CLOAD LOC	ELEMENT CLOAD LOC
	' 6	• •	6		0	0 •		9)	•))))) '

	ANGLE	-79.98 -26.71 -26.71 -26.73		Š	800.1 80.45.6 41140 800.70		ž	83.10 755.10 -5.37 -3.37 -3.37		ANGLE	2444 24626 24626 24626		9	666. 6.07.7. 6.00.00 446.7.00 13.00		ANOLE	4441 48400 68400 644 6464 66664		No.	65.14 70.14 11.18
1	Z E S	-0.10856D 05 -0.10811D 05 -0.10713D 05 -0.1085D 05		S-M	-0. 11084D 05 -0. 95790D 04 -0. 12794D 05 -0. 12109D 05 -0. 12213D 05		S-M1	-0. 12537D 05 -0. 12623D 05 -0. 12648D 05 -0. 11370D 05		NIM-8	-0.140390 05 -0.138130 05 -0.143220 05 -0.146090 05		S-H	-0.137440 05 -0.135270 05 -0.134140 05 -0.154140 05		S-MIN	-0. 12443D 05 -0. 12412D 05 -0. 12412D 05 -0. 13991D 05		S-MI	-0.113400 04 -0.113400 05 -0.113400 05 -0.122060 05
	X X X X X X X	-0.10818D 05 -0.10769D 05 -0.10836D 05 -0.10635D 05 -0.10927D 05		S-MA	-0.22694D 03 0.32925D 04 -0.41475D 04 0.14218D 04 -0.20454D 04		¥	-0. 47204D 03 -0. 20293D 03 -0. 72675D 03 -0. 24663D 03 -0. 12074D 04		S-MAX	-0.23491D 04 -0.18480D 04 -0.28833D 04 -0.22723D 04		S-MAX	-0. 57231D 04 -0. 5248BD 04 -0. 62213D 04 -0. 63887D 04 -0. 50593D 04		S-MAX	-0 87270 04 -0 887270 04 -0 936870 04 -0 936870 04 -0 803260 04		S-M-S	-0.10382D 05 -0.10439D 05 -0.10305D 05 -0.10882D 05
	215	-0.64781D 01 0.16729D 02 -0.28780D 02 -0.33493D 02 0.77612D 02		51	0. 287830 03 0. 630070 04 0. 399690 04 -0. 697420 03 0. 539950 03		215	0. 14386D 04 0. 29828D 04 0. 26271D 04 -0. 10833D 04 0. 94570D 02		S12	0.28814D 04 0.33069D 04 0.33565D 04 -0.17950D 04		515	0.26476D 04 0.26673D 04 0.27737D 04 -0.11492D 04 0.26188D 03		215	0. 13412D 04 0. 12648D 04 0. 13889D 04 -0. 26640D 02 0. 28078D 03		51	0.382970 03 0.319920 03 0.425190 03 0.251940 03
	833	-0.108550 05 -0.168430 05 -0.168460 05 -0.106340 05		533	-0.10114D 05 -0.84393D 04 -0.11897D 05 -0.90441D 04		833	-0.11575D 05 -0.11329D 05 -0.11777D 05 -0.10404D 05		833	-0.131590 05 -0.128995 05 -0.134240 05 -0.133930 05		833	-0.132550 05 -0.130430 05 -0.146830 05 -0.118290 05		833	-0.122050 05 -0.1228120 05 -0.123810 05 -0.109499 05			-0.113990 05 -0.113690 05 -0.11360 05 -0.121340 05
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	8-45-4 8-45-4	00000	95710 69090 09220 05220 63450 67220 64	-0. 36887D -0. 105887D -0. 21743D -0. 208864D -0. 20866D	Connon	138200 179320 120340 135330	nnnnn 00000	0.15346D 04 0.10189D 05 0.51486D 04 0.17412D 04 0.46248D 03	0- 0- 0- 0- 0- 0- 0- 0- 0- 0- 0- 0- 0- 0	000 B1D 48D 044 83D 044 044	-0. 196931 -0. 238061 -0. 238061 -0. 189931	nnnnn 00000	80011 40000
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	81457 81457	00000	143380 05 135170 05 136140 05 83330 04 129290 04	-0. 78171D -0. 58327D -0. 96202D -0. 19128D -0. 1424D	44400	149430 141350 156320 163990 135840	nnnnn 00000	0.60566D 04 0.5965D 04 0.50367D 04 0.59337D 04 0.24534D 04	44000 0000 00000 00000 00000	740 04 740 04 740 04 930 04 010 04	-0. 179711 -0. 168031 -0. 193571 -0. 212321 -0. 149871	nnnnn 00000	25.18 1.21.000 1.21.000 2.21.000
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LOAD LOC	118	225	833	812	S-MAX		ANOLE
22222 27774 2777		-0.109000 -0.108030 -0.108420 -0.106920 -0.113060 093		0.11941D 63 0.14543D 63 0.11015D 03 -0.96819D 02 -0.35182D 01	-0.10734b -0.10744b -0.10744b -0.10764b -0.10767b -0.10947b -0.10947b	-0.100966 -0.100970 -0.100970 -0.107720 -0.107450 -0.113060 09	800 6-180 800 6-180 800 6-180
ELEMENT (36)	8 25	833	515	S-M-R	NIH-B	ANGLE
7-4-7	-0.308310 03 0.808760 03 0.146820 03 0.516960 04 -0.417180 04	-0. 586890 03 -0. 169310 05 -0. 344160 05 -0. 365670 05 -0. 275390 05	-0.18121D 05 -0.13100D 05 -0.23801D 05 -0.18028D 05	0. 30231D 04 0. 16374D 05 0. 10900D 05 -0. 34312D 04 -0. 43276D 04	0. 2261BD 03 0. 10361D 03 -0. 98464D 04 0. 38504D 04 -0. 33961D 04	-0.31644b -0.26683b -0.39232b -0.36848b -0.28314b -0.28314b	80.81 23.78 23.92 -10.63
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ELEMENT (38)	\$25	833	812	S-HAX	N.F-2	ANOLE
2000 80777 80457	-0.137320 -0.134230 -0.168420 -0.743190 0.743190 0.43190	-0.123130 05 -0.871780 04 -0.134860 05 -0.199790 05	-0.15980D 05 -0.17152D 05 -0.17752D 05 -0.17775D 05	0, 67311D 04 0, 65392D 04 0, 65752D 04 -0, 74101D 04 -0, 35803D 04	-0. 62341D 04 -0. 83779D 04 -0. 64821D 04 -0. 54821D 04	-0.19811D 05 -0.21950D 05 -0.24047D 05 -0.16272D 05	22.1. 22.1. 22.1. 22.1. 22.1. 20.00 20.00
ELEMENT (39)	855	833	812	S-HAX	21 E-8	ANOLE
בבבבב היקרה היקרה	-0.11592D 05 -0.11830D 05 -0.13362D 05 -0.97897D 04 -0.85706D 04	-0.12107D 05 -0.12103D 05 -0.1279BD 05 -0.1379BD 05	-0.14180b 05 -0.13795b 05 -0.14620b 05 -0.13371b 05	0.38187D 04 0.38649D 04 0.36934D 04 -0.37269D 04 -0.18438D 04	-0. B000BD 04 -0. 704B7D 04 -0. B9839D 04 -0. 79473D 04	-0.15658D 05 -0.16481D 05 -0.17581D 05 -0.17581D 05	125.92 18.92 18.93 18.96 19.96
ELEMENT (40)	S25	833	SI2	S-MAX	NI H-S	ANOLE
22222 R1277	00000	-0.11709D 05 -0.10692D 05 -0.11524D 05 -0.14077D 05	-0 12708B 09-0-0-12793B 09-0-0-12793B 09-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-	0.20463D 04 0.20463D 04 0.20126D 04 -0.12940D 04 -0.92176D 03	-0.92946b 04 -0.89129b 04 -0.96771b 04 -0.96248b 04 -0.89742b 04	-0.13336b 03 -0.13717b 03 -0.14648b 03 -0.12256b 03	39.56 47.00 -19.70 -17.09
ELEMENT (41)	825	233	512	AAM-A	ZI E-S	ANGLE
22774 87774	-0.10640D 05 -0.10948D 05 -0.10748D 05	-0.113620 05 -0.109350 05 -0.112440 05 -0.126110 05	-0 11534D 05 -0 11534D 05 -0 11553D 05 -0 12385D 05	0. 78548D 03 0. 82351D 03 0. 91580D 03 -0. 55457D 03	-0.10136b 05 -0.10087b 05 -0.10168b 05 -0.10634b 05	-0.11753D 03 -0.12024D 03 -0.12754D 03 -0.12767D 03	4444 119.44 119.40 100.40 100.40

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NIE-0	-0.11024b -0.11084b -0.10972b -0.10784b -0.11338b -0.11338b	894720 09-0-178190 09-0-0-48040 09-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0-0	6-HIN -0.263450 -0.370170 -0.498910 -0.212430 -0.212430	8-HIN -0. 19890D 09 -0. 21307D 09 -0. 23240D 09 -0. 17430D 09	S-MIN -0.19235D D9 -0.19235D D9 -0.17610D 09 -0.17610D 09	S-MIN -0.133180 03 -0.132090 03 -0.136290 03 -0.126290 03	S-HIN -0.12028D 05 -0.11868D 05 -0.12207D 05
S-HAX	-0.1056350 -0.1056350 -0.1056750 -0.1036000 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.1036000 -0.1036000 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600 -0.103600	S-HAX -0.101050 -0.226490 -0.932220 -0.916690 -0.916690 -0.916690 -0.916690	89-89-87 O-1-0. 46-83-0 09-1-0. 122-750 09-1-0. 126-83-0 09-1-0. 126-83-0 09-1-0. 99-66-20 09-1-0	S-MAX -0.86712D 04 -0.78717D 04 -0.90503D 04 -0.83128D 04	S-MAX -0.93207D 04 -0.10034D 09 -0.94407D 09	S-MAX -0.98621D 04 -0.10174D 09 -0.10174D 09 -0.95681D 09	S-MAX -0.10224D 09 -0.102214D 09 -0.102745D 09
812	0.11930b 03 0.19771b 03 0.13935b 03 -0.20782b 03 -0.1810b 03	S12 0. 212650 0. 2310650 0. 231030 -0. 506060 0. 160710 0. 160710	\$12 0. 11269D 09 0. 97432D 09 0. 11042D 09 -0. 19149D 09	\$12 0. 539490 0. 538260 0. 558260 0. 560330 0. 387500 0. 387500 0. 387500 0. 387500 0. 387500 0. 387500 0. 387500 0. 387500	\$12 0.30294D 04 0.322917D 04 0.32291D 04 -0.26840D 04 -0.20377D 04	S12 0.16241D 04 0.18304D 04 -0.16701D 04 -0.1172D 04	\$12 0 662860 03 0 809150 03 0 771340 03
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\$22	-0.10886D 05 -0.10997D 05 -0.11084D 05 -0.92804D 04 -0.11489D 05	\$22 -0.19010D 05 -0.80523D 04 -0.18978D 05 -0.19028D 05	522 -0.19428D 05 -0.56374D 04 -0.19370D 09 -0.19397D 09	822 -0.19097D 09 -0.10404D 09 -0.22998D 09	522 -0.162150 09 -0.126490 09 -0.211980 09 -0.217990 09	522 -0.13734D 05 -0.1117D 05 -0.16514D 09 -0.16514D 09	S22 -0.11851D 09 -0.11879D 09 -0.11279D 09
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20	M1111 N-417	ר בי אויירה ה היידור ה היידור ה	ר בארות ה מייל איי	אויררה ר רנגעות רנגעות	רוב ארוב ה הרב היים ה	~ NULL 0 ~ NULL 0 ~ NULL 0 ~ NULL 0 N	~ 6 8 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
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-0.110872 -0.109280 -0.104640 -0.119420 -0.119420	B-MIN -0. 18037D 09 -0. 17419D 09 -0. 1829D 09 -0. 17829D 09	S-MIN -0.17921D 05 -0.1697D 05 -0.1884D 05 -0.18641D 05	S-MIN 175655 165325 165325 165326 165326 166433 164633 164633 164633 164633 164633 16463 1	S-MIN -0. 16321D 05 -0. 16683D 05 -0. 19687D 05 -0. 1987D 05	S-HIN -0.14224D 09 -0.140609D 09 -0.17009D 09 -0.17009D 09	6-1225 -0.1225 -0.1225 -0.1425 -0.14132 -0.1122 -0.012
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•	N. W. S.	10.110224 10.110424 10.110404 10.10040 1	2 E-0	-0.19203D 05 -0.20694D 05 -0.20694D 05 -0.18394D 05		S-M11 0. 17996D	-0.17467D 05 -0.18639D 05 -0.19834D 05 -0.16226D 05		8-H	-0.16551D 05 -0.17394D 05 -0.17134D 05 -0.18499D 05 -0.1476D 05		NIE-S	-0.14403D 09 -0.15364D 09 -0.16862D 09 -0.13197D 09		NIM-8	100.13322D 100.133322D 100.133324D 100.133324D 100.133325D 100.133		NIM-S	-0.12859D -0.12860D -0.128
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1	512	-0.22624b -0.19919b -0.59104b -0.31276b 0.311976b 0.311976b	SIS	-0.37871D 03 -0.68799D 04 0.91656D 03 -0.39106D 03 -0.78268D 03		S1 .49929D 0	0. 61236D 04 0. 80886D 04 -0. 60751D 03 -0. 18426D 04		212	-0. 28373D 03 0. 94353D 04 0. 62047D 04 -0. 87447D 03 -0. 22522D 04		212	0. 611320 03 0. 332200 04 0. 372300 04 -0. 241100 03 -0. 146460 04		512	0.12040D 04 0.75743D 03 0.12938D 04 0.92240D 03 -0.26126D 03		215	0. 84514D 03 -0. 63348D 03 -0. 29643D 03 0. 13153D 04
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	225	-0. 105050 -0. 107930 -0. 11630 -0. 116330 -0. 116330	S22	-0. 191950 05 -0. 281800 04 -0. 214170 04 -0. 183600 05		S2.	-0. 62207D 04 -0. 76999D 04 -0. 19831D 03 -0. 16006D 03		225	-0.16544D 05 -0.93079D 05 -0.10332D 05 -0.18433D 05 -0.14332D 05		225	-0.148350 03 -0.101460 03 -0.108790 03 -0.168340 03 -0.128410 03		822	-0.12763D 05 -0.10668D 05 -0.10481D 05 -0.15157D 09		Sas	-0.11069D 05 -0.10369D 05 -0.13018D 05
	811	-0.1111330 -0.10951D 05 -0.1061D 05 -0.1061D 05 -0.11962D 05	511	-0.90462D 03 -0.146537D 03 -0.26649D 05 -0.14893D 04 -0.35735D 03	19)	81 0. 22108D 0	-0. 141330 05 -0. 3425380 09 -0. 342240 04 -0. 782200 03	(08	811	-0. 46809D 04 -0. 11268D 03 -0. 11466D 03 -0. 66748D 04 -0. 33362D 04	81)	811	-0. 77211D 04 -0. 11891D 05 -0. 94644D 05 -0. 63666D 04	(2)	511	-0. 10730D 05 -0. 123109D 05 -0. 12811D 05 -0. 12002D 05	(2)	511	-0.12080D -0.12638D -0.12364D -0.12482D -0.53
, Jen	רסעם רסכ	XH4CF0	ELEMENT (7	ארנרט רנגייצ	ELEMENT (7	CEN	**************************************	ELEMENT (8	רסים רסכ	איירה דיינים דיינים	ELEMENT (8	LOAD LOC	**************************************	ELEMENT (8	LOAD LOC		ELEMENT (8	LOAD LOC	21-1-1 21-4-1
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Z I I I	0. 118950 10. 118950 10. 118950 10. 10. 10. 10. 10. 10. 10. 10. 10. 10.	9-411N -0. 19844D 09 -0. 21910D 09 -0. 21618D 09 -0. 18056D 09	S-MIN -0.17636D 09 -0.17784D 09 -0.20960D 09 -0.14478D 09	8-11.00 11.00 10.10.00 10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.10.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 10.00.00 1	8-100.1331260 -0.1331260 -0.1331360 -0.1341880 -0.1341880 -0.1341880 -0.099	S
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Siz	0.12322D 03 -0.42971D 03 -0.51819D 03 0.80905D 03	S12 -0.35926D 03 -0.22895D 04 -0.16159D 03 -0.93916D 03	S12 -0.11847D 03 0.47748D 04 0.90842D 04 0.28091D 03 -0.17634D 04	\$12 0.13511D 03 0.33666D 04 0.58384D 04 0.26437D 03 -0.18711D 04	\$12 0.70805D 03 0.17969D 04 0.30943D 04 0.56166D 03 -0.11325D 04	S12 0.13080D 04 0.31357D 03 0.10144D 04 0.13236D 04 0.88734D 02
Ecs	-0.123040 05 -0.123490 05 -0.123490 09 -0.133410 09 -0.13440 09	S33 -0.38767D 05 -0.33719D 05 -0.44020D 05 -0.41156D 05 -0.36451D 05	\$33 -0.342970 05 -0.316000 05 -0.393960 05 -0.293960 05	\$33 -0.28028D 05 -0.26374D 05 -0.33573D 09 -0.23072D 09	533 -0.22069D 05 -0.21090D 05 -0.26574D 05 -0.18156D 05	533 -0.17616D 05 -0.17121D 05 -0.18123D 05 -0.14682D 05
\$25 	-0, 10480D 05 -0, 10608D 05 -0, 11089D 05 -0, 11730D 05	S22 -0.19838D 05 -0.12868D 04 -0.21617D 05 -0.18010D 05	\$22 -0.176335 05 -0.885745 04 -0.778365 04 -0.142645 09	\$22 -0.15080D 05 -0.10901D 05 -0.18272D 05 -0.11851D 05	522 -0.13028D 03 -0.10786D 03 -0.15349D 04 -0.15349D 03	522 -0.11532D 05 -0.10360D 05 -0.13604D 09 -0.10727D 05
S111	-0. 115370 -0. 111080 -0. 112720 -0. 103410 -0. 118120 -0. 009	85) S11 -0. 713450 02 -0. 154810 03 -0. 154810 03 -0. 118710 04	86) 264470 04 -0. 154040 04 -0. 993420 04 0. 421570 03	87) 811 -0. 557870 04 -0. 127930 04 -0. 929030 04 -0. 232650 04	88) 811 -0. 796760 04 -0. 120390 09 -0. 108200 09 -0. 532830 04	89) -0. 10121D 03 -0. 12520D 03 -0. 11845D 03 -0. 76174D 04
רטאס רסכ	Z-CLE F-LE	ELEMENT (LOAD LOC 1 CEN 1 L-1 1 L-1 1 L-1	ELEMENT (LOAD LDC 1 CEN 1 L-1 1 L-1 1 K-L	ELEMENT C LOAD LOC ''I CEN 11 L-L-1	ELEMENT (LOAD LOC 1 CEN 1 L-1 1 L-1 1 K-L	ELEMENT (LOAD LOC 11 1-1-1 12 1-1-1 13 1-1-1 14 1-1-1 15 1-1-1 16 1-1-1 17 1-1-1 17 1-1-1 17 1-1 18 1-1 18 1-1 19

LOAD LOC	74470 11116 1247	COAD LOC LOAD LOC 1 CEN 1 L-1 1 L-1 1 K-L	ELEMENT C LOAD LDC 1 CEN 1 CEN 1 L-1	CLGAD LOC CLGAD LOC CLGAD LOC	ELEMENT COX	ELEMENT CAN LOC	ELEMENT (LOAD LOC 1 CEN
811	-0.11671D 03 -0.11638D 03 -0.10728D 03 -0.10128D 03	92) 811 -0.11732b 04 -0.13320b 04 -0.78784b 04 -0.13069b 09	93) S11 -0.38374D 04 -0.19193D 04 -0.79389D 04 -0.13835D 04	811 -0. 63023D 04 -0.38416D 03 -0.88237D 04 -0.14963D 09	95) -0.80793D 04 -0.13011D 05 -0.17992D 04 -0.99483D 04 -0.14137D 05	96) 60. 938990 04 -0. 120480 04 -0. 111210 05 -0. 134760 05	97) \$111 -0.10765D 05 -0.11276D 05
825	-0.10286b 09 -0.10243b 09 -0.10893b 09 -0.11183b 09	522 -0.14819D 05 -0.12487D 05 -0.82359D 04 -0.42918D 04	522 -0.127090 09 -0.168300 09 -0.379500 04 -0.459770 04	522 -0.105930 -0.748100 -0.790750 -0.147080 -0.106520 05520	\$22 -0.939090 -0.130260 -0.817940 -0.209210 -0.101150 -0.93	\$22 -0.991890 04 -0.124010 05 -0.951980 04 -0.428230 04 -0.945960 04	\$22 -0.997160 -0.123750 065
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. 812	0. 36290 -0. 36290 -0. 395520 0. 266670 0. 100950 0. 100950	89338D 03 0.12740D 04 0.25306D 03 0.90157D 04 0.19526D 04	S12 0.18896D 04 0.25878D 04 0.45034D 03 0.88608D 03	S12 0.28210D 04 0.36874D 04 0.91615D 03 0.45374D 04 -0.76307D 03	S12 0.30512D 04 0.39260D 04 0.10173D 04 0.2282BD 04 -0.10687D 04	S12 0.276260 04 0.350830 04 0.741850 03 0.815000 03 -0.820110 03	812 0.20327D 04 0.25170D 04
NAM-R	-0.103470 03 -0.100820 03 -0.106280 03 -0.106480 03	S-MAX -0.11149D 04 -0.35789D 04 0.13366D 04 0.96034D 03 -0.38770D 04	S-MAX -0.34517D 04 -0.84066D 04 0.15385D 04 0.94199D 04	S-HAX -0.49035D 04 -0.98172D 04 0.48912D 03 0.74935D 03	S-MAX -0.56916D 04 -0.16409D 04 -0.14770D 04 -0.98484D 04	S-MAX -0. 66910D 04 -0. 87118D 04 -0. 41865D 04 -0. 92985D 04	S-MAX -0. 82973D 04 -0. 92473D 04
	-0.11910D 05 -0.11800D 05 -0.12057D 05 -0.11244D 05	S-M1N -0.14878D 09 -0.17325D 09 -0.17675D 09 -0.13484D 09	8-130950 -0.176250 -0.76250 -0.130830 -0.140130 099	S-M1N -0.11992D 05 -0.7533D 05 -0.76124D 04 -0.10744D 09	6-119790 03 -0. 119790 03 -0. 169450 04 -0. 105650 05 -0. 144630 05	8-MIN -0.12218D 05 -0.15737D 05 -0.1217D 04 -0.11217D 05 -0.13637D 05	8 - 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
ANGLE	- 726. 98 - 66. 98 - 66. 133. 139. 100.	ANOLE 19.373 744.4033 0.334 0.334	ANGLE 17:04 12:45 73:45 6:42	ANOLE 326. 37 793. 949 949. 949 949. 949	ANOLE 38.04 44.95 74.95	ANOLE 444.43.93 89.3016 89.3016 89.3016	ANGLE 30. 52 38. 84

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	833	-0.13%690 05 -0.133750 05 -0.131580 05 -0.124290 05	S33 -0.13694D 05 -0.13661D 09 -0.13255D 09
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855	-0. 90007B 04 -0. 94273B 04 -0. 98705B 04 -0. 98705B 04	822 -0.244110 03 -0.263340 03 -0.243230 03 -0.210230 03	822 -0. 25058D 03 -0. 24582D 03 -0. 22739D 03 -0. 13663D 03	822 -0. 23760D 03 -0. 24034D 03 -0. 16908D 03 -0. 10242D 03	522 -0. 202890 05 -0. 196980 05 -0. 196980 05 -0. 103020 05 -0. 743610 04	-0.197700 03 -0.199960 03 -0.194970 03 -0.626670 04 -0.669380 04	S22 -0.116150 05
833	-0.69267D 04 -0.7213CD 04 -0.66576D 04 -0.43835D 04	533 -0.228600 -0.239140 -0.245300 -0.245300 -0.21120 0.31120	\$33 -0.193480 -0.197110 -0.193280 -0.28290 -0.161250 -0.161250	533 -0.139910 -0.140630 -0.137990 -0.173670 -0.173670 -0.173670 -0.173670 -0.173670 -0.099	633 -0. 682320 04 -0. 875480 04 -0. 875330 04 -0. 705700 05	833 833 833 833 833 833 833 833 833 833	533
215	0 33821D 01 -0.20342D 03 -0.20314D 03 -0.14603D 04	512 -0.12202D 03 -0.17752D 04 -0.4872BD 02 -0.40274D 03	512 -0. 978200 03 -0. 189260 04 -0. 249780 03 -0. 343750 03	512 -0.13971D 04 -0.22242D 04 -0.12945D 04 -0.13084D 03	512 -0.19882D 04 -0.23947D 04 -0.19082D 04 -0.26978D 04	\$12 -0.166890 04 -0.181080 04 -0.178730 04 -0.353390 04	S12 -0.104010 04
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Z1E-9	-0.90007D 04 -0.94515D 04 -0.16536D 04 -0.75299D 04	B-MIN -0.24421D 03 -0.22465D 03 -0.25721D 03 -0.2371D 03	B-HIN -0. 251160 -0. 258350 -0. 271790 -0. 230010 009	8-11N -0. 23341D 0. -0. 24331D 0. -0. 24032D 0. -0. 21830D 0.	8-HIN -0.206270 03 -0.201160 03 -0.218130 03 -0.194320 03	B 1600 1600 1600 16300 1000 16300 1000 100	S-HIN -0. 117730 05
- WGL	-6.79 -6.79 -60.97 -48.30	ANOLE -24. 39 -25. 506 -27. 506 -27. 349 -28. 389 -28.	A TAINING	ANOLE -7. 78 -12. 13 -13. 140 -83. 80 -83. 96	ANOL 119.66	800400 R	ANGL 18.6

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7-47 7-47 1111 1422	-0.639240 G4 -0.697280 G4 -0.100750 04 -0.100750 04 -0.660340 09	-0. 89503D 04 -0. 9199BD 04 -0. 90110D 04 -0. 5269BD 04	-0. 63.70%D 04 -0. 6380%D 04 -0. 6628%D 04 -0. 865699 04 -0. 40330D 04	-0.399290 03 -0.543180 03 -0.218880 03 -0.720040 03 -0.153990 04	-0.65266D 04 -0.68474D 04 -0.63017D 04 -0.8647BD 04 -0.42989D 04	-0. 90161D 04 -0. 93252D 04 -0. 88294D 04 -0. 76147D 05	6.1.4.0.0 6.1.4.0.0 6.9.4.0.0 7.9.9.0 7.9.9.0
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ELEMENT (LOAD LOC 1 CEN 1 CE	S11 -0.187830 03 -0.196190 03 -0.268860 03 -0.227230 03	522 -0.24790b 03 -0.24890b 03 -0.2244b 03 -0.13144b 03	533 -0.19379D 05 -0.19625D 05 -0.22661D 05 -0.15999D 05	S12 -0.74298D 03 -0.22408D 04 -0.79268D 03 -0.37829D 03	8-AX -0.186950 05 -0.187950 05 -0.2213350 05 -0.1250 05	8-MIN -0.2448800 05 -0.2457140 05 -0.270180 05 -0.277180 05	ANG 126 9 9 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
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ELEMENT (LOAD LOC 1 CEN 1 CEN 1 L-1 1 L-1 1 K-L	\$11 -0.927780 04 -0.983540 04 -0.216980 05 -0.186520 05	622 -0.19911D 05 -0.2014D 05 -0.10057D 05 -0.72120D 04	533 -0.907370 04 -0.919880 04 -0.108260 05 -0.732620 04	\$12 -0.30426D 04 -0.35086D 04 -0.7986D 04 -0.20333D 04	S-MAX -0.84413D 04 -0.80383D 04 -0.99764D 04 -0.68613D 04	S-MIN -0.20347D 09 -0.217657D 09 -0.21778D 09 -0.19002D 09	ANGLE 113.37 114.83 83.27 80.22
ELEMENT (LOAD LOC 1 CEN 1 1-1	\$11 -0.643930 04 -0.600860 04 -0.163930 05 -0.143940 05	S22 -0.15316D 05 -0.14764D 05 -0.63678D 04 -0.61976D 04	S33 -0. 59637D 04 -0. 60597D 04 -0. 57997D 04 -0. 54626D 04	S12 -0.2587D 04 -0.27285D 04 -0.80228D 03 -0.30624D 04	S-MAX -0. 37346D 04 -0. 347470D 04 -0. 63033D 04 -0. 51800D 04	8-MIN -0.166001D 09 -0.16693D 09 -0.16693D 09 -0.19619D 09	AN - 1111 - 1789
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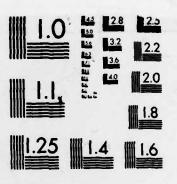
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2/2 DEVELOPMENT OF A DISTRIBUTED BREACH FOR THE CONICAL SHOCK TUBE(U) UNIVERSITY OF CENTRAL FLORIDA ORLANDO ENGINEERING AND INDUSTR.. S M METWALLI ET AL. JAN 83 TR-106-14 N00014-82-K-2049 F/G 20/11 AD-A142 210 UNCLASSIFIED NL END DATE 7-84 ptic



MICROCOPY RESOLUTION TEST CHART
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