

AD-A131 504

SINGULARITIES AT THE TIP OF A CRACK NORMAL TO THE  
INTERFACE OF AN ANISOTR. (U) ILLINOIS UNIV AT CHICAGO  
CIRCLE T C TING ET AL. MAY 83 AMMRC-TR-83-28

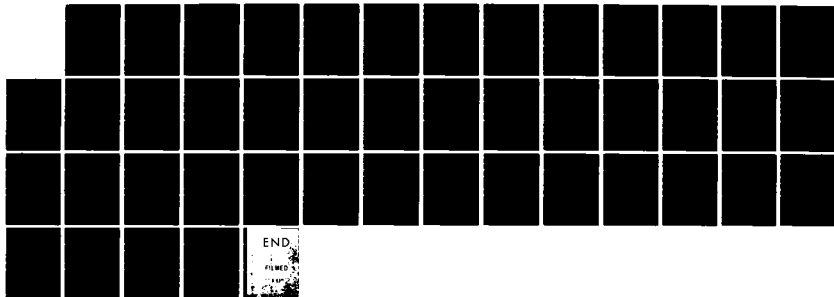
1/1

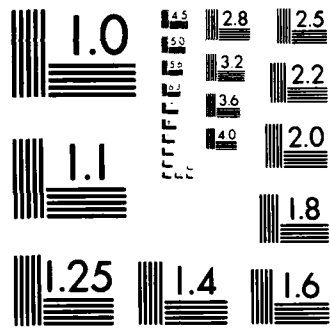
UNCLASSIFIED

DAAG46-81-K-0086

F/G 11/4

NL



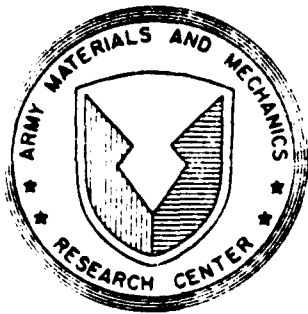


MICROCOPY RESOLUTION TEST CHART  
NATIONAL BUREAU OF STANDARDS-1963-A

AD A 131504



AD



AMMRC TR 83-28

SINGULARITIES AT THE TIP OF A CRACK NORMAL TO THE INTERFACE OF AN ANISOTROPIC LAYERED COMPOSITE

May 1983

T. C. T. TING and P. H. HOANG  
University of Illinois at Chicago  
P.O. Box 4348  
Chicago, Illinois 60680

FINAL REPORT

Contract No. DAAG46-81-K-0086

Approved for public release; distribution unlimited.

DTIC  
ELECTE  
AUG 19 1983  
S D

Prepared for

ARMY MATERIALS AND MECHANICS RESEARCH CENTER  
Watertown, Massachusetts 02172

DTIC

88 08 19 83

The findings in this report are not to be construed as an official Department of the Army position, unless so designated by other authorized documents.

Mention of any trade names or manufacturers in this report shall not be construed as advertising nor as an official indorsement or approval of such products or companies by the United States Government.

DISPOSITION INSTRUCTIONS

Destroy this report when it is no longer needed.  
Do not return it to the originator.

UNCLASSIFIED

SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)

REPORT DOCUMENTATION PAGE		READ INSTRUCTIONS BEFORE COMPLETING FORM						
1. REPORT NUMBER AMMRC TR 83-28	2. GOVT ACCESSION NO. AD-A131504	3. RECIPIENT'S CATALOG NUMBER						
4. TITLE (and Subtitle) SINGULARITIES AT THE TIP OF A CRACK NORMAL TO THE INTERFACE OF AN ANISOTROPIC LAYERED COMPOSITE		5. TYPE OF REPORT & PERIOD COVERED Final Report 1 Oct 81 to 30 Nov 82						
7. AUTHOR(s) T. C. T. Ting and P. H. Hoang		6. PERFORMING ORG. REPORT NUMBER						
9. PERFORMING ORGANIZATION NAME AND ADDRESS University of Illinois at Chicago P.O. Box 4348 Chicago, Illinois 60680		8. CONTRACT OR GRANT NUMBER(s)  DAAG46-81-K-0086						
11. CONTROLLING OFFICE NAME AND ADDRESS Army Materials and Mechanics Research Center ATTN: DRXMR-K Watertown, Massachusetts 02172		10. PROGRAM ELEMENT, PROJECT, TASK AREA & WORK UNIT NUMBERS D/A Project: 8H363304D215 AMCMS Code: 69300Q.21506						
14. MONITORING AGENCY NAME & ADDRESS (if different from Controlling Office)		12. REPORT DATE May 1983						
		13. NUMBER OF PAGES 42						
		15. SECURITY CLASS. (of this report)  Unclassified						
		15a. DECLASSIFICATION/DOWNGRADING SCHEDULE						
16. DISTRIBUTION STATEMENT (of this Report)  Approved for public release; distribution unlimited.								
17. DISTRIBUTION STATEMENT (of the abstract entered in Block 20, if different from Report)								
18. SUPPLEMENTARY NOTES								
19. KEY WORDS (Continue on reverse side if necessary and identify by block number)								
<table border="0"> <tr> <td>Composite materials</td> <td>Eigenvalues</td> </tr> <tr> <td>Stress concentration</td> <td>Anisotropy</td> </tr> <tr> <td>Complex variables</td> <td>Elasticity</td> </tr> </table>			Composite materials	Eigenvalues	Stress concentration	Anisotropy	Complex variables	Elasticity
Composite materials	Eigenvalues							
Stress concentration	Anisotropy							
Complex variables	Elasticity							
20. ABSTRACT (Continue on reverse side if necessary and identify by block number)  (SEE REVERSE SIDE)								

UNCLASSIFIED

SECURITY CLASSIFICATION OF THIS PAGE(When Data Entered)

Block No. 20

ABSTRACT

The order of stress singularities at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic layers in a composite is studied. Assuming that the stress singularities have the form  $r^{-\kappa}$ , equations are derived for determining the order of singularities  $\kappa$ . If the materials on both sides of the interface are identical,  $\kappa = \frac{1}{2}$  is a root of multiplicity three each of which can be identified with the singularity due to, respectively, a symmetric tensile stress applied at infinity, an anti-symmetric plane shear stress and an antiplane shear stress applied at infinity. When the materials on both sides of the interface are not the same, there are in general three distinct roots for  $\kappa$ . Numerical examples for a typical high modulus graphite/epoxy and for a special T300/5208 graphite/epoxy show that  $\kappa$  has three positive roots all of which are close to  $\frac{1}{2}$  for most combinations of ply-angles in the two materials. ←

UNCLASSIFIED

SECURITY CLASSIFICATION OF THIS PAGE(When Data Entered)

FORWARD

This research was performed for the Army Materials and Mechanics Research Center (AMMRC), Watertown, Massachusetts under Contract No. DAAG 46-81-K-0086 with the University of Illinois at Chicago, Chicago, Illinois. Mr. J. F. Dignam of the AMMRC was the project manager and Dr. S. C. Chou was the technical monitor. The support and encouragement of Mr. Dignam and Dr. Chou are gratefully acknowledged. We also acknowledge the assistance of Mr. Dong K. Nguyen in some of the computer calculations.

Accession For	
NTIS GRA&I	<input checked="" type="checkbox"/>
DTIC TAB	<input type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By _____	
Distribution/	
Availability Codes	
P	

11 1978

## 1. INTRODUCTION

Lightweight composite materials have been used for many years in the aerospace industries. Some of the most needed design informations are the failure mechanism and the failure criteria. Determination of the failure mechanism and the failure criteria requires in general a rigorous stress analysis at the singular point in the material. The lack of such an analysis is one of the impediments in understanding the failure mechanism and in determining the failure criteria. In the case of layered composite materials in which each layer is of orthotropic material, experimental observations indicate that the failure modes are in general either along the interface between the layers or transverse to the layers. For instance, consider the layered composite shown in Fig. 1 in which each layer is a fiber reinforced composite laminate. The fibers lie in the plane of the layers although the orientation  $\theta$  of the fibers may vary from layer to layer. When the composite is subjected to an extensional strain in the  $x_3$ -direction, a delamination may occur along the free-edge MN. The stress singularities at the free-edge such as the point M were investigated in [1]. It was shown in [1] that, except for certain special combinations of the ply-angles on both sides of the interface, the stresses have the logarithmic singularity at the free-edge in addition to the  $r^{-\kappa}$  ( $\kappa > 0$ ) singularity. Moreover, unlike the  $r^{-\kappa}$  singularity whose existence depends on the stacking sequence of the layers and the complete boundary conditions, the existence of the logarithmic singularity at point M depends only on the ply-angles on both sides of the interface. Instead of the delamination, a transverse crack shown by the dashed line may occur in the layer. There are stress singularities along the transverse crack edge MQ which is on



the interface between two layers and hence may initiate a delamination along the interface. The nature of the singularities at the crack edge such as point 0 is the main interest of this investigation.

For this purpose, we take point 0 as the origin of the rectangular coordinate system  $(x_1, x_2, x_3)$  in which the  $x_2 = 0$  plane is the interface while the transverse crack lies on the  $x_3 = 0$  plane, Fig. 1. In the  $(x_2, x_3)$  plane, Fig. 2, the  $x_3$ -axis is the interface between the two materials and a crack in material 1 is located along the negative  $x_2$ -axis. The deformation at and near the origin 0 is certainly three-dimensional, i.e., the displacements  $u_i$  are functions of  $x_2$ ,  $x_3$  as well as  $x_1$ . However, it was shown in [2] that, to the first order of approximation, the order of singularity at point 0 is affected by the dependence on  $x_1$  only to the extent of the strain component  $\epsilon_{11}$  at point 0. Since the problem is linear, we will first study the singularity at point 0 due to the displacement field in which  $u_i$  are functions of  $x_2$  and  $x_3$  only. We will then study the singularity at point 0 due to a uniform extension in the  $x_1$ -direction and see if additional singularities are present.

When the materials on both sides of the interface are isotropic, the singularity at the tip of a crack which is normal to and ends at the interface has been studied by many investigators (see [3-8], for example). In real composites, each layer is a fiber reinforced laminated material and hence should be regarded as an anisotropic material. The problem of a crack terminating at the interface of two anisotropic materials does not seem to have been investigated, although the case of orthotropic materials was studied in [9,10]. In Section 2 we present the basic formulations for anisotropic materials in two-dimensional deformation. Instead of using

Lekhnitskii's approach [11] which breaks down in degenerate cases [12] and requires a special treatment for the degenerate cases, we employ the analysis which was originally due to Stroh [13] and further developed by others [14,15] for studying surface waves in anisotropic elastic solids. In Section 3 we study the stress singularities at a crack tip in a homogeneous anisotropic elastic material, i.e., the material is not layered. The same problem was studied in [16,17] for anisotropic elastic solids whose material property is symmetric with respect to the  $x_1 = 0$  plane, and in [18,19] for general anisotropic materials. By a different approach from that of [16-19], we show that  $\kappa = 0.5$  is a triple root singularity for a crack tip in a homogeneous anisotropic medium. In Section 4 we formulate the equations for determining the order of singularities at the tip of a crack which is normal to and ends at an interface between two general anisotropic materials. We also show how one can determine the number of roots  $\kappa$  in a given region in the complex  $\kappa$ -plane. Numerical examples are presented in Section 5 in which both materials across the interface are of the same orthotropic materials although the orientations of the axes of symmetry are different. It is shown that there are three positive roots for  $\kappa$  and that all three have the values between 0.333 and 0.698. The fact that we have three positive  $\kappa$  and that they are near the value 0.5 is not surprising in view of the results obtained in Section 3 for the crack in a homogeneous anisotropic medium. Finally, in Section 6 we study the stress distribution near the tip of the crack which is normal to the interface subject to a uniform extensional strain in the  $x_1$ -direction. We show that a uniform stress solution exists and hence no additional stress singularity is present due to the uniform extension.

## 2. ANISOTROPIC MATERIAL IN TWO-DIMENSIONAL DEFORMATION

In the rectangular coordinate system  $(x_2, x_3)$ , let the displacements  $u_i$  ( $i = 1, 2, 3$ ) and hence the strains  $\epsilon_{ij}$  and the stresses  $\sigma_{ij}$  be functions of  $x_2$  and  $x_3$  only. The strain-displacement, stress-strain and equilibrium equations can be written as

$$\epsilon_{ij} = (\partial u_i / \partial x_j + \partial u_j / \partial x_i) / 2 \quad (1)$$

$$\sigma_{ij} = c_{ijkl} \epsilon_{kl} \quad (2)$$

$$\partial \sigma_{i2} / \partial x_2 + \partial \sigma_{i3} / \partial x_3 = 0 \quad (3)$$

where repeated indices imply summation and

$$c_{ijkl} = c_{jikl} = c_{kpij} \quad (4)$$

are the elasticity constants. A general solution for Eqs. (1,2,3) can be obtained by letting

$$u_i = u_i f(Z) \quad (5)$$

$$Z = x_2 + px_3 \quad (6)$$

where  $p$  and  $u_i$  are constants to be determined and  $f$  is an arbitrary function of  $Z$  [13-15]. Substituting into Eqs. (1,2,3) we obtain

$$\sigma_{ij} = \tau_{ij} df/dZ \quad (7)$$

$$D_{ik} u_k = 0 \quad (8)$$

where

$$\tau_{ij} = (c_{ijk2} + pc_{ijk3}) u_k \quad (9)$$

$$D_{ik} = c_{i2k2} + p(c_{i2k3} + c_{i3k2}) + p^2 c_{i3k3} \quad (10)$$

For a non-trivial solution of  $u_i$ , it follows from Eq. (8) that the determinant of  $D_{ik}$  must vanish. That is,

$$\|D_{ik}\| = 0 \quad (11)$$

This results in a sextic equation for  $p$ . Since the eigenvalues  $p$  are all non-real [11,13], there are three pairs of complex conjugates for  $p$  which will be denoted by  $p_L$  and  $\bar{p}_L$  ( $L=1,2,3$ ) where an over bar denotes the complex conjugate, and three pairs of associated eigenvectors  $u_{i,L}$  and  $\bar{u}_{i,L}$  ( $L=1,2,3$ ). It should be pointed out that some elements of  $\tau_{ij}$  given by Eq. (9) are related to each other. We can show from Eqs. (4), (8) and (10) that  $\tau_{ij} = \tau_{ji}$  and

$$\left. \begin{aligned} \tau_{12} &= -p\tau_{13} , \\ \tau_{22} &= -p\tau_{23} = p^2\tau_{33} \end{aligned} \right\} \quad (12)$$

They agree with Lekhnitskii's results in [11] except that  $\tau_{33}$  is taken as unity in [11]: However,  $\tau_{33}$  may vanish in degenerate cases [12] and hence Lekhnitskii's approach requires a special treatment whenever  $\tau_{33} = 0$ .

The general solution for the displacements and stresses can now be written as

$$u_i = \sum_L \{u_{i,L} f_L(z_L) + \bar{u}_{i,L} g_L(\bar{z}_L)\} \quad (13)$$

$$\sigma_{ij} = \sum_L \{ \tau_{ij,L} df_L(z_L)/dz_L + \bar{\tau}_{ij,L} dg_L(\bar{z}_L)/d\bar{z}_L \} \quad (14)$$

where the summation is over  $L=1,2,3$  and  $f_L$  and  $g_L$  are, respectively, arbitrary functions of  $z_L$  and  $\bar{z}_L$ . Notice that Eq. (13) contains six arbitrary functions. When  $p$  has a double root, the solution given by Eq. (13) is not general unless one has two independent  $u_i$  associated with the double root  $p$ . Similarly, Eq. (13) may not be a general solution when  $p$  is a triple root. A modified solution which maintains its generality

when  $p$  has a multiple root was given in [20]. For isotropic materials,  $p = i$  and is a triple root. However, one of the triple roots is associated with the anti-plane deformation while the other two which can be considered a double root are associated with plane strain deformations [21].

To study the singularities at a crack tip, we let

$$\left. \begin{aligned} f_L(Z_L) &= A_L Z_L^{1-\kappa} / (1-\kappa) \\ g_L(\bar{Z}_L) &= B_L \bar{Z}_L^{1-\kappa} / (1-\kappa) \end{aligned} \right\} \quad (L \text{ not summed}) \quad (15)$$

where  $\kappa$ ,  $A_L$  and  $B_L$  are arbitrary constants which may be complex. Introducing the polar coordinates  $(r, \phi)$ :

$$x_2 = r \cos \phi, \quad x_3 = r \sin \phi \quad (16)$$

Equations (13) and (14) can be written as

$$u_i = r^{1-\kappa} \sum_L \left\{ A_L u_{i,L} \zeta_L^{1-\kappa} + B_L \bar{u}_{i,L} \bar{\zeta}_L^{1-\kappa} \right\} / (1-\kappa) \quad (17)$$

$$\sigma_{ij} = r^{-\kappa} \sum_L \left\{ A_L \tau_{ij,L} \zeta_L^{-\kappa} + B_L \bar{\tau}_{ij,L} \bar{\zeta}_L^{-\kappa} \right\} \quad (18)$$

where

$$\zeta_L = \cos \phi + p_L \sin \phi \quad (19)$$

We see that if the real part of  $\kappa$  is positive,  $\sigma_{ij}$  is singular at  $r = 0$ .

When  $\kappa$  is real, one may choose without loss of generality [1]

$$B_L = \bar{A}_L = (a_L + i\tilde{a}_L) / 2 \quad (20)$$

where  $a_L$  and  $\tilde{a}_L$  are real constants. Equations (17) and (18) then have the real expressions

$$u_i = r^{1-\kappa} \sum_L \left\{ a_L \operatorname{Re} \left( u_{i,L} \zeta_L^{1-\kappa} \right) + \tilde{a}_L \operatorname{Im} \left( u_{i,L} \zeta_L^{1-\kappa} \right) \right\} / (1-\kappa) \quad (21)$$

$$\sigma_{ij} = r^{-\kappa} \sum_L \left\{ a_L \operatorname{Re}(\tau_{ij,L} \zeta_L^{-\kappa}) + \tilde{a}_L \operatorname{Im}(\tau_{ij,L} \zeta_L^{-\kappa}) \right\} \quad (22)$$

where  $\operatorname{Re}$  and  $\operatorname{Im}$  stand for real and imaginary, respectively.

We may also use  $c_{ij}$  instead of  $c_{ijkl}$  and write Eqs. (2) and (4) as

$$\sigma_i = c_{ij} \varepsilon_j, \quad c_{ij} = c_{ji} \quad (23)$$

where

$$\left. \begin{aligned} \sigma_1 &= \sigma_{11}, & \sigma_2 &= \sigma_{22}, & \sigma_3 &= \sigma_{33} \\ \sigma_4 &= \sigma_{23}, & \sigma_5 &= \sigma_{13}, & \sigma_6 &= \sigma_{12} \end{aligned} \right\} \quad (24a)$$

$$\left. \begin{aligned} \varepsilon_1 &= \varepsilon_{11}, & \varepsilon_2 &= \varepsilon_{22}, & \varepsilon_3 &= \varepsilon_{33} \\ \varepsilon_4 &= 2\varepsilon_{23}, & \varepsilon_5 &= 2\varepsilon_{13}, & \varepsilon_6 &= 2\varepsilon_{12} \end{aligned} \right\} \quad (24b)$$

We will also write the inverse of Eq. (23) as

$$\varepsilon_i = s_{ij} \sigma_j, \quad s_{ij} = s_{ji} \quad (25)$$

where  $s_{ij}$  are the elastic compliances.

### 3. SINGULARITIES AT A CRACK TIP IN A HOMOGENEOUS ANISOTROPIC MATERIAL

In this section we assume that material 1 and material 2 in Fig. 2 are the same material. Therefore, there is no interface and the crack is in a homogeneous medium. This problem has been studied in [16,17] where the material property is assumed to be symmetric with respect to the  $x_1 = 0$  plane. No symmetry is assumed in this section. Sih and Chen [18] and Hoenig [19] also studied the problem without assuming the symmetry, but the analysis presented here will show that  $\kappa = \frac{1}{2}$  is a root of multiplicity three. The boundary conditions at the crack surface are

$$\sigma_{3j} = 0 \quad \text{at} \quad \phi = \pm\pi \quad (26)$$

From Eq. (19) we have

$$\zeta_L = e^{\pm\pi i} \quad \text{at} \quad \phi = \pm\pi \quad (27)$$

and application of Eq. (26) to (18) yields

$$\sum_L \left\{ A_L \tau_{3j,L} e^{-\kappa\pi i} + B_L \bar{\tau}_{3j,L} e^{\kappa\pi i} \right\} = 0 \quad (28)$$

$$\sum_L \left\{ A_L \tau_{3j,L} e^{\kappa\pi i} + B_L \bar{\tau}_{3j,L} e^{-\kappa\pi i} \right\} = 0 \quad (29)$$

where  $j = 1, 2, 3$ . Equations (28,29) may be written in matrix notation as

$$\underline{K}(\kappa)\underline{q} = \underline{0} \quad (30)$$

where  $\underline{q}$  is a column matrix whose elements are  $A_L$  and  $B_L$  ( $L = 1, 2, 3$ ), while  $\underline{K}$  is a 6x6 square matrix and is a function of  $\kappa$ . For a nontrivial solution of  $\underline{q}$  we must have

$$\|\underline{K}(\kappa)\| = 0 \quad (31)$$

which provides the roots  $\kappa$ .

Notice that when  $\kappa = \frac{1}{2}$ ,  $e^{\kappa\pi i} = -e^{-\kappa\pi i} = i$  and the left-hand side of Eq. (28) is identical to that of Eq. (29) except the minus signs. Hence the first three rows of  $\underline{K}$  are identical to the last three rows except the signs and the rank of  $\underline{K}$  is no more than 3. This means that  $\kappa = \frac{1}{2}$  is a root of multiplicity of at least 3 and that there are at least 3 independent solutions for  $\underline{q}$  in Eq. (30) associated with  $\kappa = \frac{1}{2}$ .

To find  $\underline{q}$ , i.e.,  $A_L$  and  $B_L$  for  $\kappa = \frac{1}{2}$ , let

$$\sum_L \tau_{3j,L} \tau_{3s,L}^* = J \delta_{js}, \quad J = \|\tau_{3j,L}\| \quad (32)$$

where  $\delta_{js}$  is the Kronecker delta. Thus regarding  $\tau_{3j,L}$  as a 3x3 matrix,  $\tau_{3j,L}^*$  is the adjoint matrix and  $J$  is the determinant. Assuming that  $J \neq 0$ , the solution given by

$$A_L = \frac{1}{2\sqrt{2}} J^{-1} k_s \tau_{3s,L}^*, \quad B_L = \frac{1}{2\sqrt{2}} \bar{J}^{-1} k_s \bar{\tau}_{3s,L}^* \quad (33)$$

satisfies Eqs. (28,29) when  $\kappa = \frac{1}{2}$ .  $k_s$  ( $s = 1, 2, 3$ ) in Eq. (33) are arbitrary real constants. With Eq. (33), Eq. (18) can now be written as, noticing that  $\kappa = \frac{1}{2}$ ,

$$\sigma_{ij}(r, \phi) = \frac{1}{(2r)^{\frac{1}{2}}} \sum_L \operatorname{Re} \left\{ J^{-1} \tau_{3s,L}^* \tau_{ij,L} \zeta_L^{-\frac{1}{2}} \right\} k_s \quad (34)$$

Equation (34) provides three independent solutions for the stress distribution near the crack tip. Notice that at  $\phi = 0$ , Eq. (34) for  $i = 3$  has the expression, after using Eqs. (19) and (32),

$$\sigma_{3j}(r, 0) = \frac{1}{(2r)^{\frac{1}{2}}} k_j, \quad j = 1, 2, 3 \quad (35)$$

Hence  $k_j$  are the stress intensity factors. It is shown in the Appendix that the solution associated with  $k_3$  can be identified with the solution near the tip of a crack length  $2a$  due to a uniform symmetric tensile stress  $\sigma_{33}$  at infinity. Similarly, the solutions associated with  $k_1$  and  $k_2$  can be identified with the solutions near the tip of a crack length  $2a$  due to uniform antisymmetric shear stresses  $\sigma_{31}$  and  $\sigma_{32}$ , respectively, at infinity.

Equation (34) is not valid when  $J = 0$ . However,  $J = 0$  implies that  $\bar{J} = 0$  and the rank of  $K$  is no more than two. It follows that  $\kappa = \frac{1}{2}$  is a root of multiplicity of at least four when  $J = 0$ . This is not possible



because for the special case of a crack in an isotropic material it is known that  $\kappa = \frac{1}{2}$  is a root of multiplicity three.

#### 4. SINGULARITIES AT THE TIP OF A CRACK NORMAL TO AN INTERFACE

We will now proceed to determine the singularities at the tip of a crack which is normal to the interface between two anisotropic materials as shown in Fig. 2. Equations (17) and (18) for the displacements and the stresses apply to materials 1 and 2. To distinguish notations in material 2 from those in material 1, we will add a prime to all notations referring to material 2 except  $\kappa$  which is the same in both materials and  $r$  and  $\phi$  which have unambiguous definitions. Since material 1 is divided into two parts by a crack, the displacements and the stresses in both parts need not be given by the same expressions. We will therefore use a superscript (+) to denote quantities referred to material 1 in the  $x_3 > 0$  region and a superscript (-) to denote quantities referred to material 1 in the  $x_3 < 0$  region.

The stress-free boundary conditions at the crack surface are

$$\left. \begin{aligned} \sigma_{3j}^+ &= 0 & \text{at } \phi &= \pi \\ \sigma_{3j}^- &= 0 & \text{at } \phi &= -\pi \end{aligned} \right\} \quad (36a)$$

and the continuity conditions at the interface are

$$\left. \begin{aligned} u_i^+ - u_i' &= 0 \\ \sigma_{2j}^+ - \sigma_{2j}' &= 0 \end{aligned} \right\} \quad \text{at } \phi = \pi/2 \quad (36b)$$

$$\left. \begin{aligned} u_i' - u_i^- &= 0 \\ \sigma_{2j}' - \sigma_{2j}^- &= 0 \end{aligned} \right\} \quad \text{at } \phi = -\pi/2 \quad (36c)$$

where  $i, j = 1, 2, 3$ . Use of the expressions for  $u_i$  and  $\sigma_{ij}$  from Eqs. (17) and (18) in Eqs. (36a), (36b), (36c) results in 18 equations for the eighteen coefficients  $A_L^+$ ,  $B_L^+$ ,  $A_L'$ ,  $B_L'$ ,  $A_L^-$  and  $B_L^-$  ( $L = 1, 2, 3$ ) which can again be written in the form

$$\underline{K}(\kappa)\underline{q} = \underline{0} \quad (37)$$

where  $\underline{K}$  is now an  $18 \times 18$  square matrix and the elements of  $\underline{q}$  are  $A_L^+$ ,  $B_L^+$ ,  $A_L'$ ,  $B_L'$ ,  $A_L^-$  and  $B_L^-$  ( $L = 1, 2, 3$ ). For a nontrivial solution of  $\underline{q}$ , we must have

$$\|\underline{K}(\kappa)\| = 0 \quad (38)$$

It should be pointed out that for real  $\kappa$  one could use the real expressions for  $u_i$  and  $\sigma_{ij}$  in Eqs. (21) and (22). We again obtain 18 equations in the form of Eq. (37) in which the elements of  $\underline{q}$  are  $a_L^+$ ,  $\tilde{a}_L^+$ ,  $a_L'$ ,  $\tilde{a}_L'$ ,  $a_L^-$  and  $\tilde{a}_L^-$ . The elements of the  $18 \times 18$  matrix  $\underline{K}$  are real.

The roots  $\kappa$  of Eq. (38), whether real or complex, can be found numerically when the elasticity constants  $c_{ij}$  and  $c'_{ij}$  of both materials are known. In the complex plane  $\kappa$ , one can find the number of roots  $\kappa$  in a given region by using the following theorem [22]: If a function  $\psi(\kappa)$ , whose only singularities inside a closed contour  $C$  of the  $\kappa$ -plane are poles and whose value is not zero at any point on the contour, then

$$\frac{1}{2\pi i} \int_C \frac{d\psi/d\kappa}{\psi} d\kappa = N - P \quad (39)$$

where  $N$  is the number of zeros and  $P$  is the number of poles inside  $C$ .

In our problem  $\psi = \|\underline{K}\|$  and Eq. (39) can be written as

$$\frac{1}{2\pi i} \int_C d\{\ln \|\underline{K}\|\} = N \quad (40)$$

because  $\|\underline{k}\|$  is bounded and hence there is no pole. In view of the fact that

$$\ln \psi = \ln |\psi| + i \arg \psi \quad (41)$$

Equation (40) reduces to

$$N = \frac{1}{2\pi} \left( \arg \|\underline{k}\| \right)_C \quad (42)$$

where  $(f)_C$  denotes the change in the value of  $f$  in going around the contour  $C$ . Equation (42) implies that if  $\|\underline{k}\|$  has  $N$  roots inside the contour  $C$  in the  $\kappa$ -plane, the locus of  $\|\underline{k}\|$  when plotted on the complex  $\|\underline{k}\|$  plane encircles the origin  $N$  times as  $\kappa$  goes around the contour  $C$ . This is a useful result which can be employed to determine the number of roots in a given region.

## 5. NUMERICAL EXAMPLES

For numerical examples we will assume that each layer in the composite shown in Fig. 1 is an orthotropic material with respect to the  $(\hat{x}_1, \hat{x}_2, \hat{x}_3)$  axes where  $\hat{x}_2 = x_2$  and the  $\hat{x}_3$ -axis is the fiber direction which makes an angle  $\theta$  with the  $x_3$ -axis. We further assume that all layers are identical orthotropic materials although the ply-angle  $\theta$  may vary from layer to layer. Referring to the  $(\hat{x}_1, \hat{x}_2, \hat{x}_3)$  axes, the following engineering constants for two different orthotropic materials are taken from [25] and [24], respectively.

### Composite W

(Typical high modulus graphite/epoxy, [23])

$$\left. \begin{aligned} E_1 &= E_2 = 2.1 \times 10^6 \text{ psi} \\ E_3 &= 20 \times 10^6 \text{ psi} \\ G_{12} &= G_{23} = G_{31} = 0.85 \times 10^6 \text{ psi} \\ \nu_{21} &= \nu_{31} = \nu_{32} = 0.21 \end{aligned} \right\} \quad (43)$$

## Composite T

(T300/5208 graphite/epoxy, [24])

$$\left. \begin{aligned}
 E_1 &= E_2 = 1.54 \times 10^6 \text{ psi} \\
 E_3 &= (20 \times 10^6 \text{ psi})^* \\
 G_{12} &= G_{23} = G_{31} = 0.81 \times 10^6 \text{ psi} \\
 \nu_{21} &= \nu_{31} = \nu_{32} = 0.28
 \end{aligned} \right\} \quad (44)$$

In Eqs. (43) and (44),  $E_i$  are the Young's moduli,  $G_{ij}$  the shear moduli and  $\nu_{ij}$  are the Poisson's ratios [25]. They are related to  $s_{ij}$  [25,26], and hence  $s_{ij}$  can be computed. Since  $E_{ij}$ ,  $G_{ij}$  and  $\nu_{ij}$  given in Eqs. (43) and (44) are referred to the  $(\hat{x}_1, \hat{x}_2, \hat{x}_3)$  axes,  $s_{ij}$  so obtained will be denoted by  $\hat{s}_{ij}$ . Its inverse  $\hat{c}_{ij}$  is obtained by using the equations derived in [25]. Finally, we obtain  $s_{ij}$  and  $c_{ij}$  which are referred to the  $(x_1, x_2, x_3)$  axes from  $\hat{s}_{ij}$  and  $\hat{c}_{ij}$  (see [2,12]) when the ply-angle  $\theta$  in material 1 is given. For a different ply-angle  $\theta'$  in material 2, we obtain  $s'_{ij}$  and  $c'_{ij}$  in the same way.

With the elasticity constants  $c_{ij}$  and  $c'_{ij}$  associated with the ply-angles  $\theta$  and  $\theta'$  in materials 1 and 2 so determined, the formulation in Section 4 provides the matrix  $\underline{K}$  and the roots of the determinant  $\|\underline{K}\|$  furnishes the desired roots. In Table 1 we list the roots  $\kappa$  for the Composite W given by Eq. (43) for combinations of  $(\theta/\theta')$  angles in which both  $\theta$  and  $\theta'$  assume the values between  $-90^\circ$  and  $90^\circ$  with an increment of  $15^\circ$  degrees. Since the values of  $\kappa$  for  $(-a/b)$  and  $(a/-b)$  composites are identical, it is sufficient to consider  $\theta > 0$  only. In Table 2 we

---

\*The value of  $E_3$  in [24] is  $22 \times 10^6$  psi. The calculations presented in Table 2 and Figs. 4 and 6 were inadvertently based on  $E_3 = 20 \times 10^6$  psi. However, a few sample calculations with the correct  $E_3$  show that the errors on  $\kappa$  are no more than 1%.

list the roots  $\kappa$  for composite T. In both tables,  $\kappa = \frac{1}{2}$  is a triple root when  $\theta = \theta'$  and when  $\theta = -\theta' = 90^\circ$ . This is expected since when  $\theta = \theta'$  and  $\theta = -\theta' = 90^\circ$  material 1 and material 2 have the same fiber orientation and hence there is no real interface between material 1 and material 2. Thus the crack is in a homogeneous material and the analysis in Section 3 shows that  $\kappa = \frac{1}{2}$  is a triple root. For all other combinations of  $(\theta/\theta')$  there are three non-equal roots of positive  $\kappa$ .

Notice that the values of  $\kappa$  are identical for  $(\theta/90)$  and  $(\theta/-90)$ . Likewise,  $\kappa$ 's are identical for  $(0/\theta')$  and  $(0/-\theta')$  as well as for  $(90/\theta')$  and  $(90/-\theta')$ . Some of the roots are not easy to locate. For instance, two of the three roots for  $(0/90)$  in Composite T differ by only  $10^{-5}$ . At the first calculation we only located the root  $\kappa = 0.697726$  and completely missed the other two roots. Great effort and care were exercised in locating the two roots which are very close to each other. We also used Eq. (42) to verify numerically that there are indeed three positive roots for  $\kappa$ .

The results presented in Tables 1 and 2 are shown in graphs in Figs. 3-6. In Fig. 3 contour lines for the largest  $\kappa$ , denoted by  $\kappa_1$ , are shown for Composite W and the corresponding contour lines for Composite T is shown in Fig. 4. In Figs. 5 and 6 the contour lines for  $(\kappa_1 - \kappa_3)$  where  $\kappa_3$  is the smallest of the three roots are shown for Composite W and Composite T, respectively. In view of symmetry, only half of the contour lines are shown in Figs. 3-6. In both composites we see that the largest  $\kappa_1$  occurs at  $(0/-90)$  which is identical to  $(0/90)$ . At  $(0/90)$ ,  $\kappa_1 = 0.676635$  for Composite W and  $\kappa_1 = 0.697726$  for Composite T. The largest difference between  $\kappa_1$  and  $\kappa_3$  occurs approximately at  $(30/-40)$  for both Composite W and Composite T.

## 6. UNIFORM EXTENSION

We will consider in this section the problem in which the composite shown in Fig. 2 is subject to a uniform extensional strain  $\epsilon_1$  in the  $x_1$ -direction. For this purpose, we add the term  $\epsilon_1 \delta_{i1} x_1$  to the right-hand sides of Eqs. (5), (13), (17) and (21) for the displacements  $u_i$  and the term  $c_{ij11} \epsilon_1$  to Eqs. (7), (14), (18) and (22) for the stresses  $\sigma_{ij}$  [1,27]. With these modifications, application of the boundary and interface conditions, Eqs. (36), results in the following system of equations

$$r^{-\kappa} \underline{K}(\kappa) \underline{q} = \epsilon_1 \underline{b} \quad (45)$$

where the elements of  $\underline{q}$  are  $A_L^+$ ,  $B_L^+$ ,  $A_L'$ ,  $B_L'$ ,  $A_L^-$  and  $B_L^-$  while  $\underline{b}$  contains  $c_{ij11}$ . Since the right-hand side of Eq. (45) is independent of  $r$ , we let  $\kappa = 0$  and obtain

$$\underline{K}(0) \underline{q} = \epsilon_1 \underline{b} \quad (46)$$

Equation (46) provides  $\underline{q}$  if  $\underline{K}(0)$  is not singular. If  $\underline{K}(0)$  is singular, a solution may still exist if  $\underline{b}$  is orthogonal to the left eigenvectors of  $\underline{K}(0)$ . However, when  $\kappa = 0$  Eq. (18) indicates that  $\sigma_{ij}$  are constants and the stresses are uniform. Therefore, instead of solving Eq. (46), we will use an alternate approach employed in [1] to find a uniform stress solution which satisfies the prescribed uniform extensional strain  $\epsilon_1$ .

To this end, we first solve  $\sigma_1$  from the first of Eq. (25):

$$\sigma_1 = (\epsilon_1 / s_{11}) - R_j \sigma_j, \quad (j \neq 1) . \quad (47)$$

We then eliminate  $\sigma_1$  from the rest of Eq. (25) to obtain

$$\epsilon_i = R_{ij} \sigma_j + R_i \epsilon_1, \quad (i \neq 1, j \neq 1) \quad (48)$$

where

$$\left. \begin{aligned} R_i &= s_{i1}/s_{11} \\ R_{ij} &= s_{ij} - s_{i1}R_j \end{aligned} \right\} \quad (49)$$

The free surface conditions, Eq. (36a), imply that

$$\left. \begin{aligned} \sigma_3^+ &= \sigma_4^+ = \sigma_5^+ = 0 \\ \sigma_3^- &= \sigma_4^- = \sigma_5^- = 0 \end{aligned} \right\} \quad (50)$$

in the entire region occupied by material 1 because the stresses are uniform.

The continuity conditions across the interface  $\phi = \pm\pi/2$  as given by

Eqs. (36b,c) can be shown to be equivalent to

$$\left. \begin{aligned} \varepsilon_1^+ &= \varepsilon_1' = \varepsilon_1^-, & \varepsilon_3^+ &= \varepsilon_3' = \varepsilon_3^-, & \varepsilon_5^+ &= \varepsilon_5' = \varepsilon_5^- \\ \sigma_2^+ &= \sigma_2' = \sigma_2^-, & \sigma_4^+ &= \sigma_4' = \sigma_4^-, & \sigma_6^+ &= \sigma_6' = \sigma_6^- \end{aligned} \right\} \quad (51)$$

From Eqs. (50) and (51), the stresses  $\sigma_i^+$ ,  $\sigma_i^-$  and  $\sigma_i'$  in the regions  $\pi/2 < \phi < \pi$ ,  $-\pi < \phi < -\pi/2$  and  $-\pi/2 < \phi < \pi/2$ , respectively, have the following non-zero components:

$$\left. \begin{aligned} \sigma_i^+ &= (\sigma_1^+, \sigma_2, 0, 0, 0, \sigma_6) \\ \sigma_i^- &= (\sigma_1^-, \sigma_2, 0, 0, 0, \sigma_6) \\ \sigma_i' &= (\sigma_1', \sigma_2, \sigma_3', 0, \sigma_5', \sigma_6) \end{aligned} \right\} \quad (52)$$

where the superscripts +, - and a prime are omitted for  $\sigma_2$  and  $\sigma_6$  in view of Eq. (51). Using Eq. (52) in Eq. (48), the expressions  $\varepsilon_3$  and  $\varepsilon_5$  in material 1 are

$$\left. \begin{aligned} \varepsilon_3 &= R_{32}\sigma_2 + R_{36}\sigma_6 + R_3\varepsilon_1 \\ \varepsilon_5 &= R_{52}\sigma_2 + R_{56}\sigma_6 + R_5\varepsilon_1 \end{aligned} \right\} \quad (53)$$

while the expressions for  $\epsilon_3$  and  $\epsilon_5$  in material 2 are

$$\left. \begin{aligned} \epsilon_3 &= R'_{32}\sigma_2 + R'_{33}\sigma_3 + R'_{35}\sigma_5 + R'_{36}\sigma_6 + R'_3\epsilon_1 \\ \epsilon_5 &= R'_{52}\sigma_2 + R'_{53}\sigma_3 + R'_{55}\sigma_5 + R'_{56}\sigma_6 + R'_5\epsilon_1 \end{aligned} \right\} \quad (54)$$

Again, the superscripts +, - and a prime for  $\epsilon_3$  and  $\epsilon_5$  are omitted in view of Eq. (51).

For the composite considered in Eq. (43) and (44), the material property in each layer is symmetric with respect to the  $x_2 = 0$  plane and hence [24,25],

$$s_{14} = s_{16} = s_{24} = s_{26} = s_{34} = s_{36} = s_{54} = s_{56} = 0 \quad (55)$$

This implies that  $R'_{36} = R'_{56} = R_{36} = R_{56} = 0$  and elimination of  $\epsilon_3$  and  $\epsilon_5$  from Eqs. (53) and (54) reduces to

$$\left. \begin{aligned} (R'_{32} - R_{32})\sigma_2 + R'_{33}\sigma_3 + R'_{35}\sigma_5 &= -(R'_3 - R_3)\epsilon_1 \\ (R'_{52} - R_{52})\sigma_2 + R'_{53}\sigma_3 + R'_{55}\sigma_5 &= -(R'_5 - R_5)\epsilon_1 \end{aligned} \right\} \quad (56)$$

Equations (56) provide a one-parameter family of solutions for  $\sigma_2$ ,  $\sigma_3'$  and  $\sigma_5'$ .  $\sigma_6$  is arbitrary while  $\sigma_1^+$ ,  $\sigma_1^-$  and  $\sigma_1'$  are determined from Eq. (47). This completes the solution for non-zero elements of the stresses shown in Eq. (52). It is not difficult to see from Eqs. (52) and (47) that  $\sigma_1^+ = \sigma_1^-$ .

Equations (56) have no solutions if

$$\frac{R'_{32} - R_{32}}{R'_{52} - R_{52}} = \frac{R'_{33}}{R'_{53}} = \frac{R'_{35}}{R'_{55}} \neq \frac{R'_3 - R_3}{R'_5 - R_5} \quad (57)$$

For the composites considered here, Eqs. (57) do not hold and a uniform stress solution exists for a given extensional strain  $\epsilon_1$ . Hence, there is no additional stress singularity at the crack tip due to the uniform extension for the composites considered here. If Eqs. (57) held, one would have a logarithmic stress singularity at the crack tip [1].



## 7. CONCLUDING REMARKS

We have shown that the stress singularity at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic media has the form  $r^{-\kappa}$  where  $\kappa$  is a root of Eq. (38). We show that there are in general three distinct positive roots of  $\kappa$ , say  $\kappa_s$  ( $s=1,2,3$ ). For each  $\kappa_s$ , Eq. (37) furnishes the associated  $q_s$  whose elements are  $A_L, B_L$  in Eq. (18). Since  $q_s$  is unique up to a multiplicative constant, say  $c_s$ , the solution given by Eq. (18) for  $\kappa = \kappa_1, \kappa_2$  and  $\kappa_3$  can be superimposed and written as

$$\sigma_{ij} = c_1 r^{-\kappa_1} \sigma_{ij}^{(1)} + c_2 r^{-\kappa_2} \sigma_{ij}^{(2)} + c_3 r^{-\kappa_3} \sigma_{ij}^{(3)} \quad (58)$$

where  $\sigma_{ij}^{(s)}$  ( $s=1,2,3$ ) are known functions of  $\phi$ . The analysis presented here provides  $\kappa_s$  and  $\sigma_{ij}^{(s)}$  ( $s=1,2,3$ ). The unknown constants  $c_1, c_2$  and  $c_3$  have to be determined from the complete boundary conditions of the whole composite. One could use the solution obtained here to form a special finite element at the crack tip and use regular finite elements elsewhere to find a complete solution numerically for the stress distribution in the composite with a crack.

It should be pointed out that Eq. (38) provides, besides the three positive  $\kappa$  presented here, infinitely many negative  $\kappa$  whose real part is negative. Thus to increase the accuracy in the special finite element at the crack tip, one may add as many terms as one pleases to the right-hand side of Eq. (58). These additional terms are non-singular at the tip of the crack.

**BLANK PAGE**

## REFERENCES

- [1] Zwierns, R. I., Ting, T. C. T. and Spilker, R. L., "On the Logarithmic Singularity of Free-Edge Stress in Laminated Composites Under Uniform Extension," *J. Appl. Mech.*, Vol. 49, Sept. 1982, pp. 561-569.
- [2] Zwierns, R. I. and Ting, T. C. T., "Free-Edge and Contact-Edge Singularities in Laminated Composites," AMMRC TR 82-21, Dept. of Materials Engineering, Univ. of Illinois at Chicago Circle, April 1982.
- [3] Zak, A. K. and Williams, M. L., "Crack Point Stress Singularities at a Bi-Material Interface," *J. Appl. Mech.*, Vol. 30, 1963, 142-143.
- [4] Bogy, D. B., "On the Plane Elastostatic Problem of a Loaded Crack Terminating at a Material Interface," *J. Appl. Mech.*, Vol. 38, 1971, 911-918.
- [5] Cook, T. S. and Erdogan, F., "Stresses in Bonded Materials with a Crack Perpendicular to the Interface," *Int. J. Engineering Science*, Vol. 10, 1972, 677-697.
- [6] Gupta, G. D., "A Layered Composite with a Broken Laminate," *Int. J. Solids Structures*, Vol. 9, 1973, 1141-1154.
- [7] Erdogan, F. and Bakioglu, M., "Fracture of Plates Which Consists of Periodic Dissimilar Strips," *Int. J. Fracture*, Vol. 12, 1976, 71-84.
- [8] Lin, K. Y. and Mar, J. W., "Finite Element Analysis of Stress Intensity Factors for Cracks at a Bi-Material Interface," *Int. J. Fracture*, Vol. 12, No. 4, 1976, 521-531.
- [9] Arin K., "A Note on the Fracture of Laminated Composites," *Letters in Applied and Engineering Sciences*, Vol. 3, 1975, 81-85.
- [10] Delale, F. and Erdogan, F., "Bonded Orthotropic Strips with Cracks," *Int. J. Fracture*, Vol. 15, 1979, 343-364.
- [11] Lekhnitskii, S. G., "Theory of Elasticity on an Anisotropic Elastic Body," (translated by P. Fern), Holden Day, San Francisco, 1963.
- [12] Ting, T. C. T. and Chou, S. C., "Stress Singularities in Laminated Composites," in *Fracture of Composite Materials*, G. C. Sih and V. P. Tamuzs, eds., Martinus Nijhoff, 1982, 265-277.
- [13] Stroh, A. N., "Steady State Problems in Anisotropic Elasticity," *J. Math. Phys.*, Vol. 41, 1962, pp. 77-103.
- [14] Barnett, D. M. and Lothe, J., "Line Force Loadings on Anisotropic Half-Space and Wedges," *Phys. Norv.*, Vol. 8, 1975, 13-22.
- [15] Chadwick, P. and Smith, G. D., "Foundations of the Theory of Surface Waves in Anisotropic Elastic Materials," *Adv. Appl. Mech.*, Vol. 17, 1977, 303-376.

- [16] Sih, G. C., Paris, P. C. and Irwin, G. R., "On Cracks in Rectilinear Anisotropic Bodies," *Int. J. Fracture Mechanics*, Vol. 1, 1965, 189-302.
- [17] Sih, G. C. and Liebowitz, "Mathematical Theories of Brittle Fracture," in Fracture, Vol. II, ed. by H. Liebowitz, Academic Press, 1968, 115-125.
- [18] Sih, G. C. and Chen, E. P., "Cracks in Composite Materials," Martinus Nijhoff Pub., 1981, 87-97.
- [19] Hoenig, Alan, "Near-Tip Behavior of a Crack in a Plane Anisotropic Elastic Body," *Eng. Fracture Mechanics*, Vol. 16, 1982, 393-403.
- [20] Ting, T. C. T., "Effects of Change of Reference Coordinates on the Stress Analyses of Anisotropic Elastic Materials," *Int. J. Solids and Structures*, Vol. 18, 1982, 139-152.
- [21] Ting, T. C. T. and Chou, S. C., "Edge Singularities in Anisotropic Composites," *Int. J. Solids and Structures*, Vol. 17, 1981, 1057-1068.
- [22] Phillips, E. G., "Functions of a Complex Variable," Oliver and Boyd, New York, 1961, 107-108.
- [23] Wang, A. S. D. and Crossman, F. W., "Some New Results on Edge Effects in Symmetric Composite Laminates," *J. Composite Materials*, Vol. 11, 1977, 92-106.
- [24] Chou, S. C., "Delamination of T300/5208 Graphite/Epoxy Laminates," in Fracture of Composite Materials, G. C. Sih and V. P. Tamuzs, eds., Martinus Nijhoff, 1982, 247-263.
- [25] Jones, R. M., "Mechanics of Composite Materials," McGraw-Hill, 1975.
- [26] Christensen, R. M., "Mechanics of Composite Materials," Wiley-Interscience, 1979.
- [27] Zwiwers, R. I. and Ting, T. C. T., "Singularity of Contact-Edge Stress in Laminated Composites Under Uniform Extension," *J. Composite Materials*, Vol. 17, 1983, 49-63.

## APPENDIX

Consider a crack of length  $2a$  located on the  $x_2$ -axis between  $x_2 = -a$  and  $x_2 = a$  in an infinite anisotropic elastic material subjected to a uniform tensile stress  $\sigma_{33}^{\infty}$  as well as a uniform plane shear stress  $\sigma_{32}^{\infty}$  (Fig. 7) and a uniform antiplane shear stress  $\sigma_{31}^{\infty}$  (not shown in Fig. 7) at infinity. The surface of the crack is stress-free. This problem was studied in [16,17] in which the material property is assumed to be symmetric with respect to the  $x_1 = 0$  plane and in [18,19] for general anisotropic materials. We will present an alternate expression for the solution in the entire plane, not just near the crack tip as in [19]. For the present problem, it suffices to consider the associated problem in which the crack surface is subjected to the stresses

$$\sigma_{3j} = -\sigma_{3j}^{\infty} \quad \text{on} \quad -a < x_2 < a \quad (\text{A1})$$

and the stresses vanish at infinity. To this end, we choose in Eqs. (13,14)

$$f_L = \bar{g}_L = A_L \left\{ (Z_L^2 - a^2)^{1/2} - Z_L \right\} / 2 \quad (\text{A2})$$

where  $A_L$  are arbitrary complex constants. We then have the solution for the displacements

$$u_i = \sum_L \operatorname{Re} \left\{ A_L u_{i,L} \left( (Z_L^2 - a^2)^{1/2} - Z_L \right) \right\} \quad (\text{A3})$$

and the stresses

$$\sigma_{ij} = \sum_L \operatorname{Re} \left\{ A_L \tau_{ij,L} \left( \frac{Z_L}{(Z_L^2 - a^2)^{1/2}} - 1 \right) \right\} \quad (\text{A4})$$

which vanish at infinity. To determine  $A_L$ , we apply the boundary conditions, Eq. (A1):

$$-\sigma_{3j}^{\infty} = \sum_L \operatorname{Re} \left\{ A_L \tau_{3j,L} \left( \frac{x_2}{\pm i(a^2 - x_2^2)^{1/2}} - 1 \right) \right\}, \quad -a < x_2 < a \quad (\text{A5})$$

Equation (A5) is satisfied if we let

$$\sum_L \tau_{3j,L} A_L = \sigma_{3j}^{\infty} \quad (\text{A6})$$

or

$$A_L = J^{-1} \sigma_{3s}^{\infty} \tau_{3s,L}^* \quad (\text{A7})$$

where  $\tau_{3s,L}^*$  is defined in Eq. (32). We therefore obtain the solution

$$u_i = \sum_L \operatorname{Re} \left\{ J^{-1} \tau_{3s,L}^* u_{i,L} \left( (Z_L^2 - a^2)^{1/2} - Z_L \right) \right\} \sigma_{3s}^{\infty} \quad (\text{A8})$$

$$\sigma_{ij} = \sum_L \operatorname{Re} \left\{ J^{-1} \tau_{3s,L}^* \tau_{ij,L} \left( \frac{Z_L}{(Z_L^2 - a^2)^{1/2}} - 1 \right) \right\} \sigma_{3s}^{\infty} \quad (\text{A9})$$

To find the stress distribution near the crack tip, say  $x_2 = a$ , notice that as  $Z_L - a \cong 0$

$$\frac{Z_L}{(Z_L^2 - a^2)^{1/2}} - 1 = \frac{(Z_L - a) + a}{(Z_L - a)^{1/2} (Z_L + a)^{1/2}} - 1 \cong \left( \frac{a}{2} \right)^{1/2} (Z_L - a)^{-1/2} \quad (\text{A10})$$

If we let  $Z_L - a = r\zeta_L$  and

$$k_s = \sqrt{a} \sigma_{3s}^{\infty} \quad (\text{A11})$$

Equation (A9) reduces to Eq. (34). Thus the solution associated with  $k_3$  in Eq. (34) is due to the uniform symmetric tensile stress  $\sigma_{33}^{\infty}$  applied at infinity while that associated with  $k_1$  and  $k_2$  are due to, respectively, the uniform anti-symmetric shear stresses  $\sigma_{31}^{\infty}$  and  $\sigma_{32}^{\infty}$  applied at infinity. It should be pointed out that the solution associated with  $\sigma_{33}^{\infty}$  is not necessarily symmetric with respect to the  $x_2$  axis even though the loading at the infinity is. Likewise, the solutions associated with  $\sigma_{31}^{\infty}$  and  $\sigma_{32}^{\infty}$  are not necessarily anti-symmetric with respect to the  $x_2$  axis.

$\theta \backslash \theta'$	0	15°	30°	45°	60°	75°	90°
90°	.676635 .500212 .5	.676012 .502156 .500222	.672441 .505068 .500236	.660488 .504813 .500227	.630471 .502297 .500165	.567537 .500314 .500048	.5
75°	.675051 .500189 .433918	.665411 .500305 .446234	.646234 .500401 .466560	.616354 .500415 .485037	.571159 .500269 .496300	.5	.499859 .499688 .432561
60°	.665263 .499839 .379767	.642118 .500352 .404849	.605628 .500735 .441810	.557636 .500673 .477316	.5	.503704 .499085 .430024	.498608 .497668 .371402
45°	.643963 .499012 .369508	.606474 .500092 .407075	.556388 .500639 .456342	.5	.522801 .498293 .443947	.515052 .496531 .388842	.495691 .495008 .345965
30°	.609394 .498562 .391694	.557954 .499986 .442491	.5	.543655 .498192 .445399	.558434 .495380 .400299	.533668 .493226 .364378	.494600 .492252 .339747
15°	.560499 .499279 .438704	.5	.557552 .498554 .444039	.592813 .495627 .400073	.595139 .492603 .369244	.553905 .490548 .349787	.497659 .489828 .340358
0	.5	.562178 .498904 .441309	.609785 .496201 .397243	.631434 .493110 .367864	.620457 .490626 .350305	.566078 .489298 .342401	.5 .489038 .341108
-15°	.560499 .499279 .438704	.612345 .497112 .392438	.644542 .494473 .364451	.651651 .492080 .349195	.628243 .490464 .342609	.564345 .489824 .341721	.497659 .489828 .340358
-30°	.609394 .498562 .391684	.645821 .496706 .361115	.662510 .494839 .347946	.656327 .493276 .344694	.621931 .492378 .346998	.552727 .492163 .349080	.494600 .492252 .339747
-45°	.643963 .499012 .369508	.665377 .497882 .352528	.668725 .496774 .351288	.651255 .495887 .357999	.608061 .495504 .367486	.538511 .495550 .368791	.495691 .495008 .345965
-60°	.665263 .499839 .379767	.674013 .499331 .373635	.666989 .498824 .381746	.641375 .498455 .396175	.592892 .498378 .410685	.526260 .498488 .409137	.498608 .497668 .371402
-75°	.675051 .500189 .433918	.674923 .500080 .435943	.663730 .499972 .447918	.638493 .499903 .462126	.593334 .499878 .474130	.523618 .499848 .476682	.499859 .499688 .432561
-90°	.676635 .500212 .5	.676012 .502156 .500222	.672441 .505068 .500236	.660488 .504813 .500227	.630471 .502297 .500165	.567537 .500314 .500048	.5

Table 1 Values of  $\kappa$  for the  $r^{-\kappa}$  singularity in Composite W

$\theta \backslash \theta'$	0	15°	30°	45°	60°	75°	90°
90°	.697726 .5 .499990	.695390 .501422 .499970	.687400 .503731 .499959	.670249 .504020 .500004	.636110 .502055 .500055	.570493 .500028 .499923	.5
75°	.694552 .499899 .433166	.682182 .499923 .446108	.658683 .500018 .465858	.623468 .500173 .484789	.573597 .500205 .496796	.5	.500075 .499786 .429903
60°	.681455 .499252 .380013	.655864 .499572 .405261	.615433 .500057 .440977	.562256 .500366 .476460	.5	.503193 .499057 .428481	.498302 .497881 .367957
45°	.655675 .498169 .370945	.615621 .499132 .408413	.561314 .500048 .456362	.5	.523662 .498551 .440550	.515275 .496552 .385476	.496507 .496224 .341936
30°	.615974 .497892 .393350	.561933 .499414 .443807	.5	.543669 .498709 .441797	.559222 .496016 .394947	.534343 .493379 .359462	.495858 .491924 .334337
15°	.562542 .499066 .439371	.5	.556290 .499047 .441155	.591495 .496372 .395197	.594666 .493174 .363520	.553999 .490735 .343852	.498878 .489680 .333397
0	.5	.561734 .499106 .439862	.608589 .496752 .393682	.630402 .493751 .362960	.620348 .491033 .344491	.566821 .489363 .335554	.5 .488944 .333540
-15°	.562542 .499066 .439371	.614806 .497105 .391399	.646999 .494775 .361050	.654617 .492462 .344079	.632281 .490645 .335970	.568975 .489711 .333613	.498878 .489680 .333397
-30°	.615974 .497892 .393350	.653467 .496339 .360418	.670723 .494834 .344095	.665388 .493407 .338249	.632110 .492360 .338301	.562294 .491864 .339280	.495858 .491924 .334337
-45°	.655675 .498169 .370945	.678402 .497416 .350987	.682644 .496660 .345625	.666150 .495899 .348752	.623644 .495381 .355521	.551195 .495156 .357113	.496507 .496244 .341936
-60°	.681455 .499252 .380013	.691616 .499019 .370260	.685470 .498713 .373343	.660390 .498389 .383591	.611548 .498218 .395570	.539832 .498183 .396061	.498302 .497881 .367957
-75°	.694552 .499899 .433166	.695563 .499852 .431371	.684047 .499759 .438689	.657219 .499696 .450101	.609624 .499704 .461274	.537075 .499720 .463714	.500075 .499786 .429903
-90°	.697726 .5 .499990	.695390 .501422 .499970	.687400 .503731 .499959	.670249 .504020 .500004	.636110 .502055 .500055	.570493 .500028 .499923	.5

Table 2 Values of  $\kappa$  for the  $r^{-k}$  singularity in Composite T



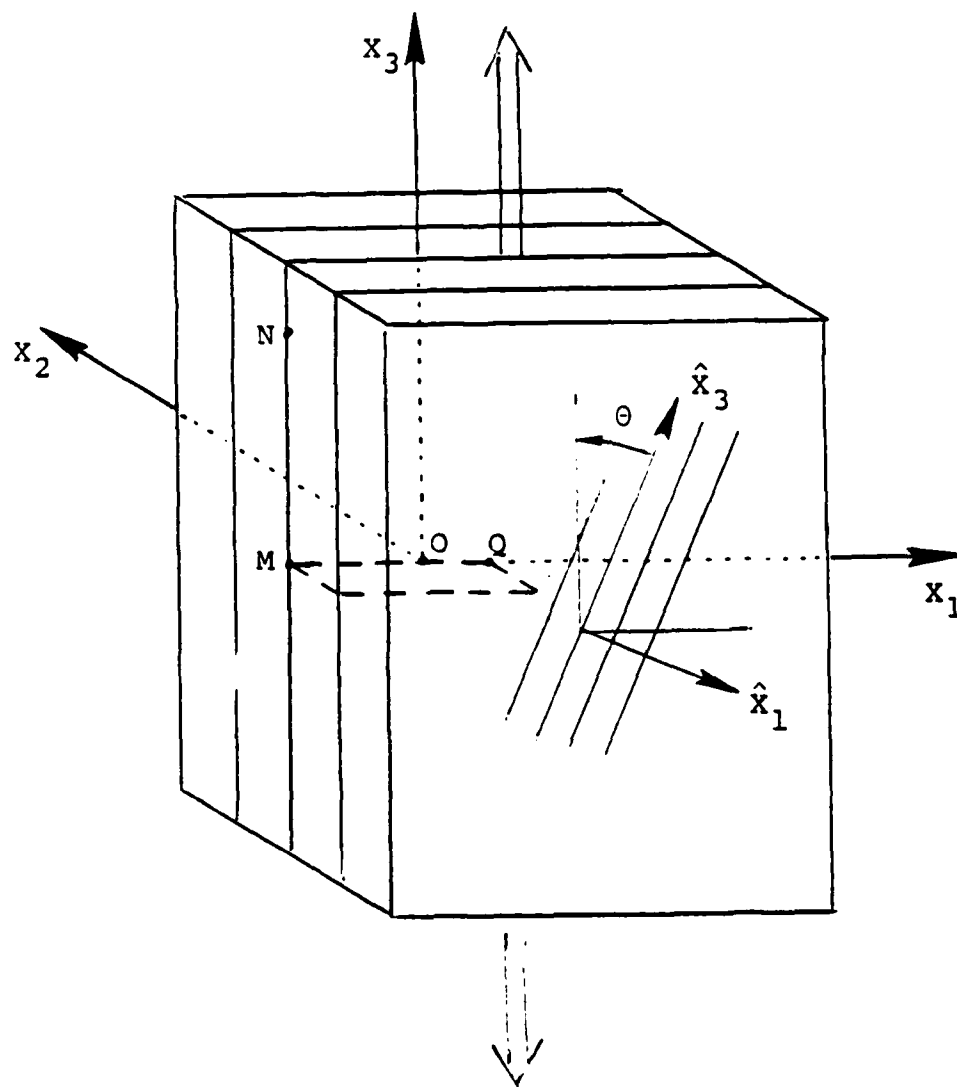


Fig. 1 Geometry of an angle-ply laminated composite

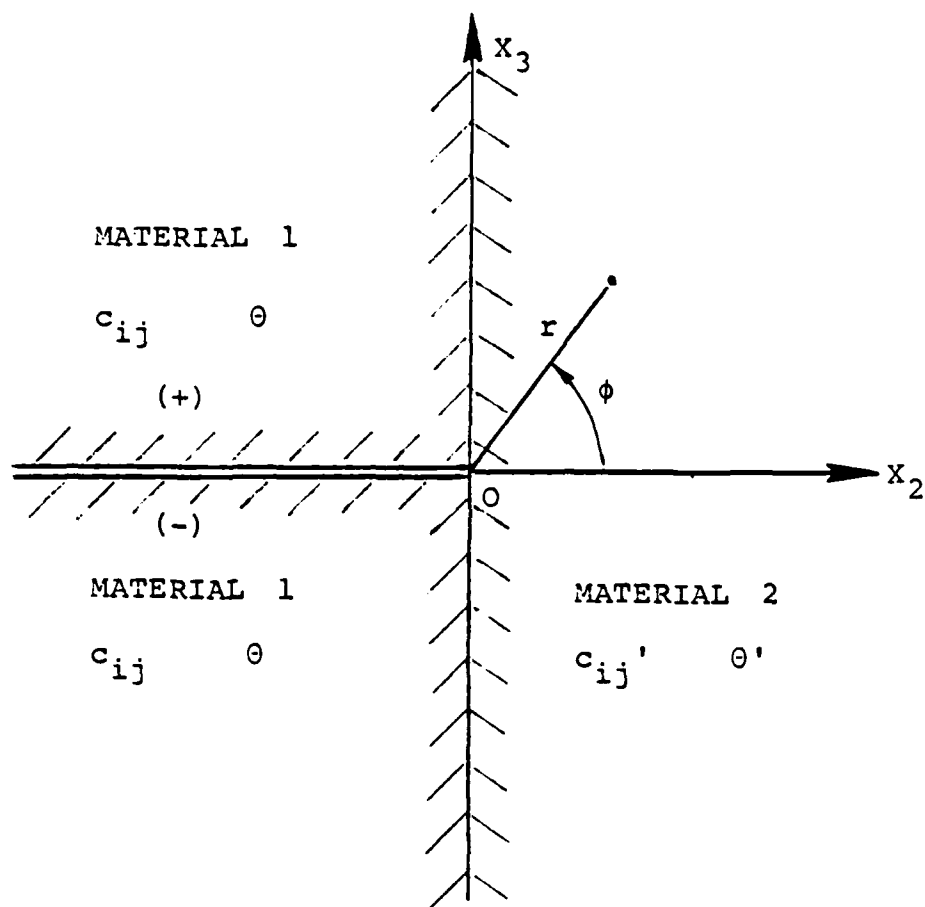


Fig. 2 A crack which is normal to and ends at the interface between two anisotropic materials

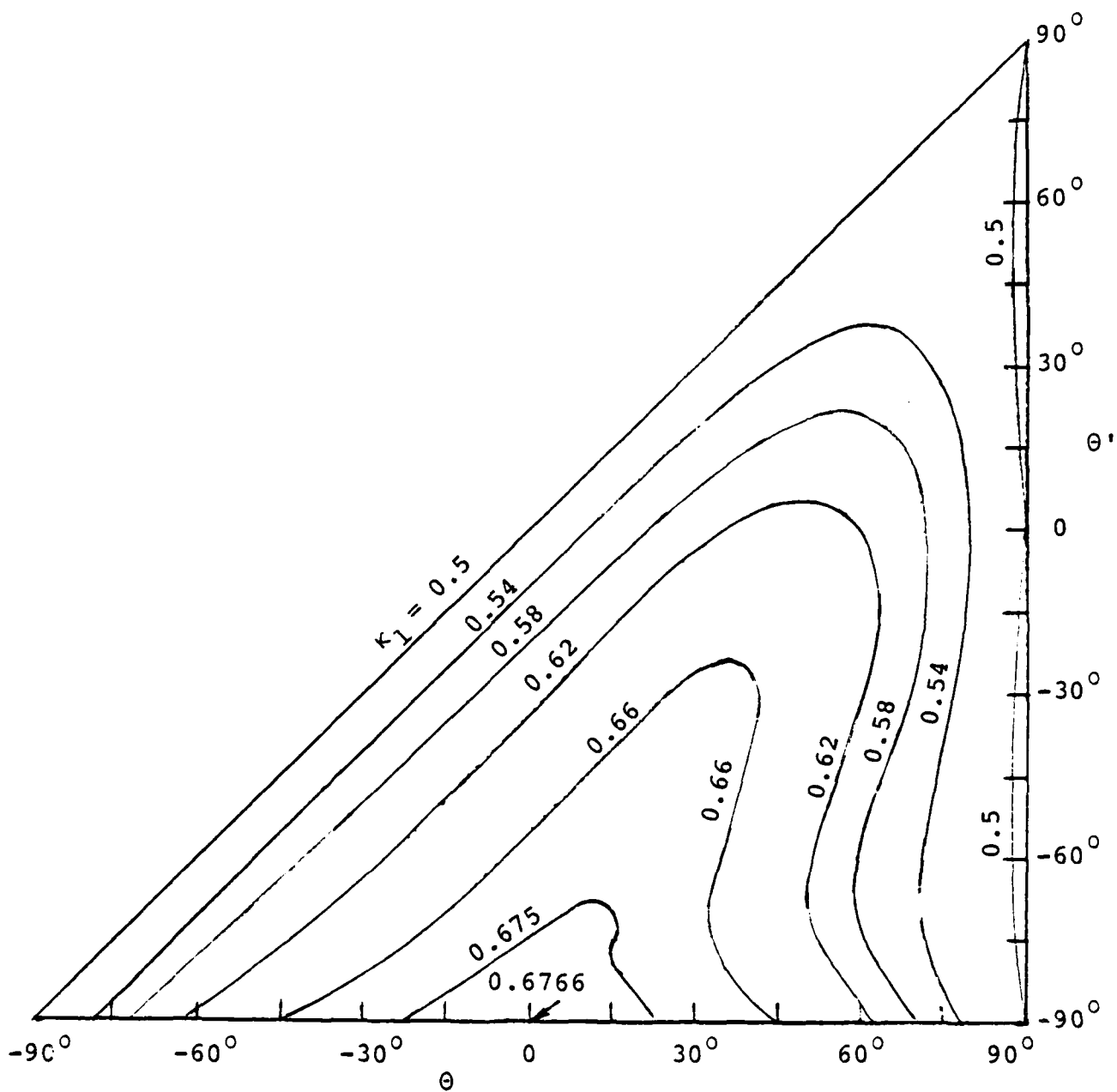


Fig. 3 Contour lines for  $\kappa_1$ , the largest  $\kappa$  of the  $r^{-\kappa}$  singularity for Composite W

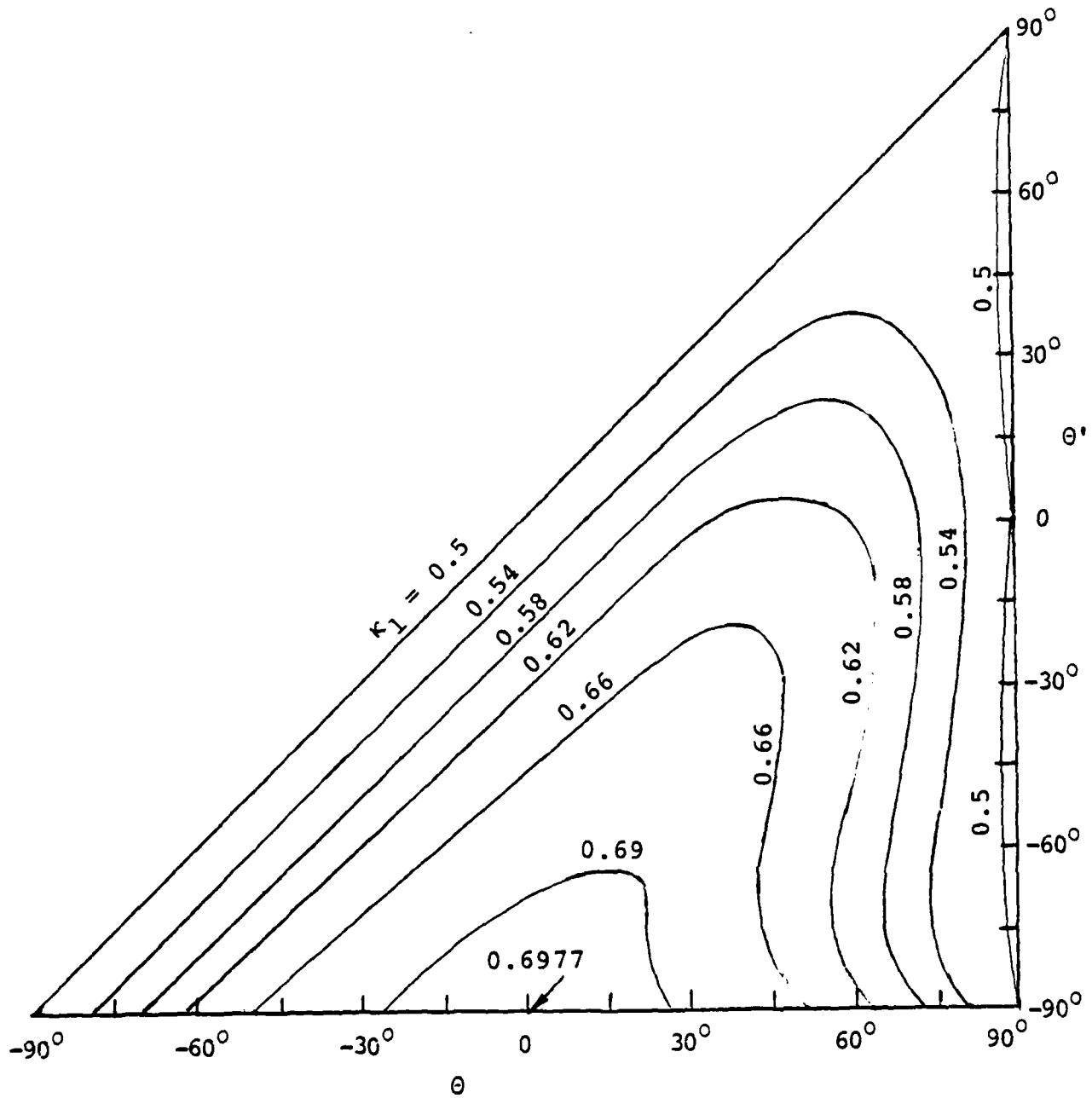


Fig. 4 Contour lines for  $\kappa_1$ , the largest  $\kappa$  of the  $r^{-\kappa}$  singularity for Composite T

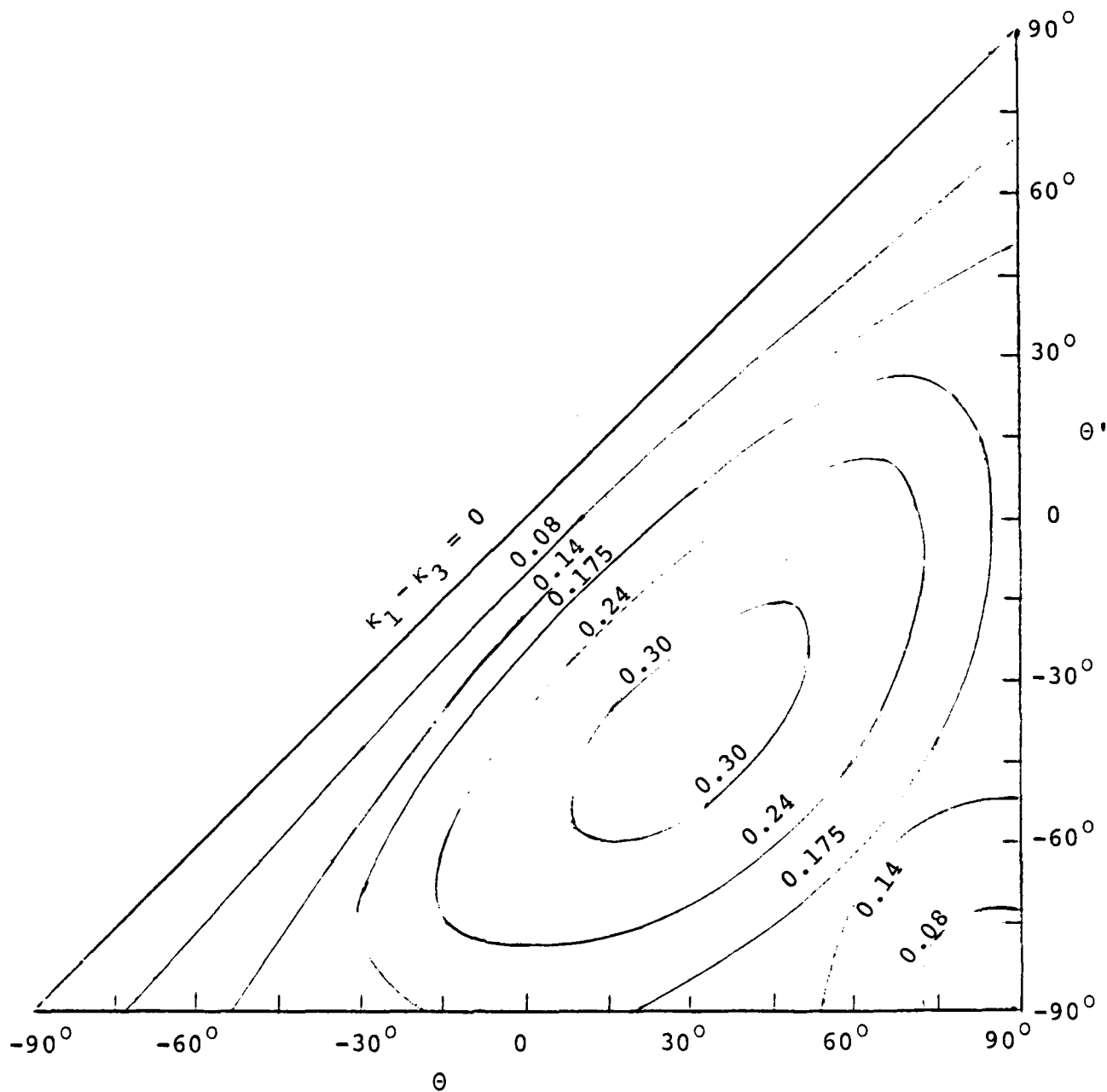


Fig. 5 Contour lines for the difference between the largest ( $\kappa_1$ ) and the smallest ( $\kappa_3$ ) singularities for Composite W

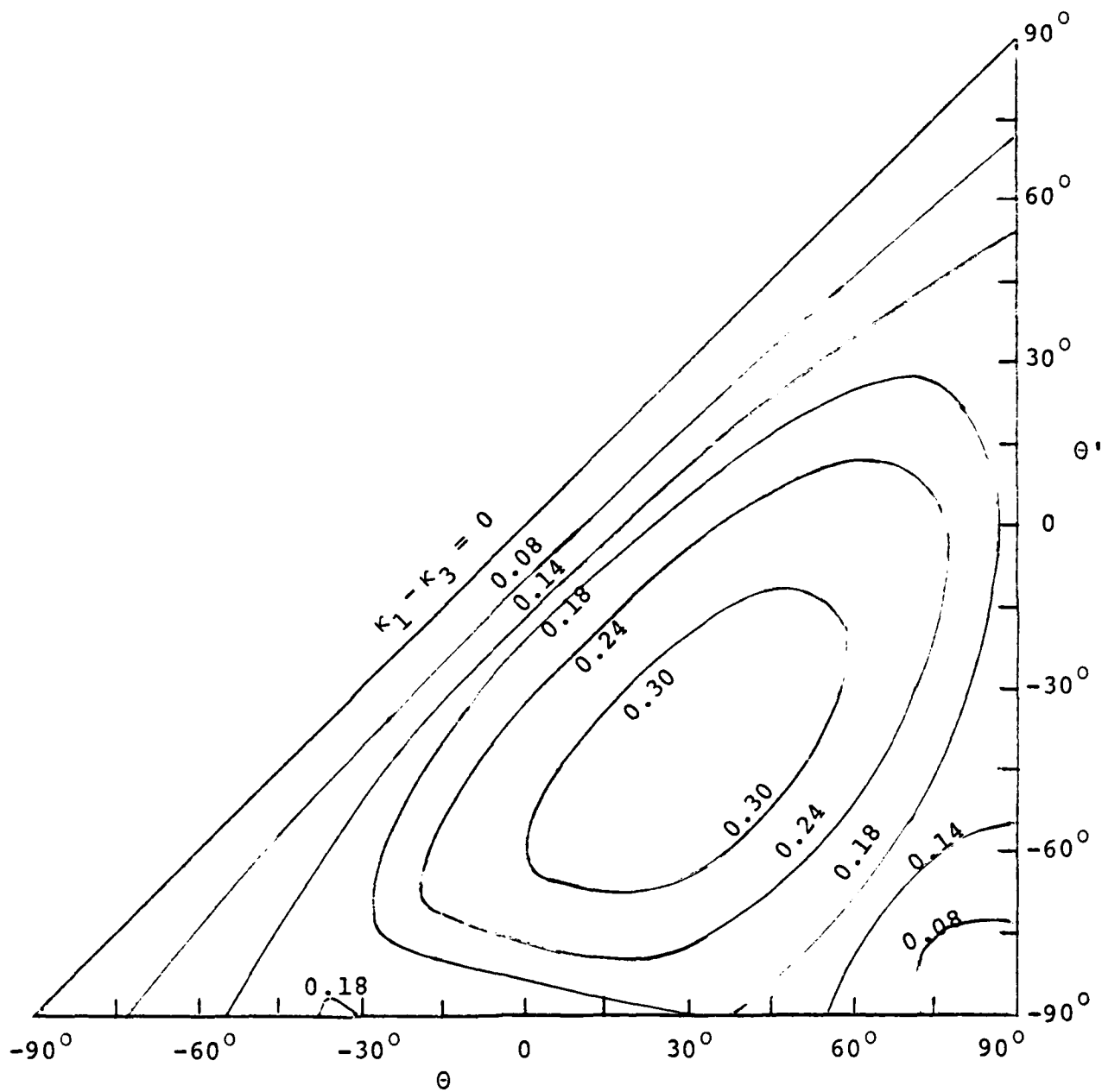


Fig. 6 Contour lines for the difference between the largest ( $\kappa_1$ ) and the smallest ( $\kappa_3$ ) singularities for Composite T

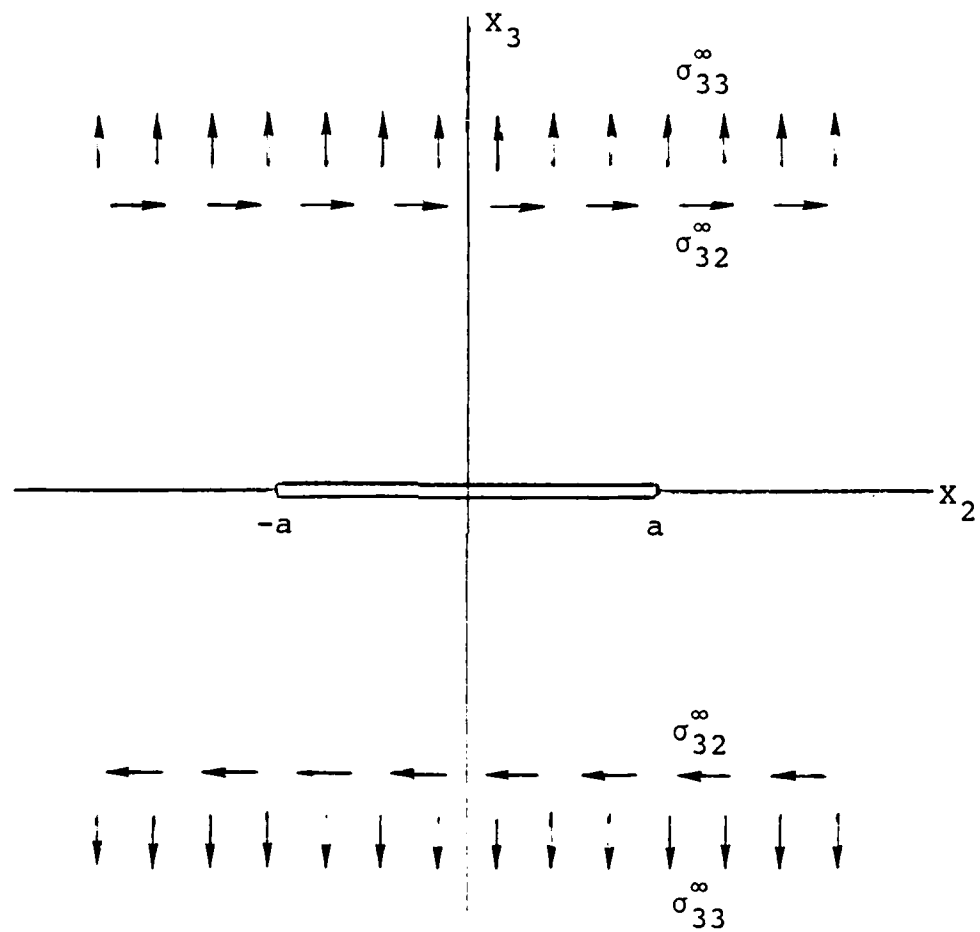


Fig. 7 A crack of length  $2a$  subject to uniform stresses at infinity

DISTRIBUTION LIST

	<u>No. of Copies</u>
Office of Deputy Under Secretary of Defense for Research and Engineering (ET) ATTN: Mr. J. Persh, Staff Specialist for Materials and Structures (Room 3D1089) The Pentagon Washington, DC 20301	1
Office of Deputy Chief of Research Development and Acquisition ATTN: DAMA-CSS/Dr. J. I. Bryant (Room 3D424) The Pentagon Washington, DC 20301	1
Commander U.S. Army Materiel Development and Readiness Command ATTN: DRCLDC, R. Gonano, Office of Laboratory Management 5000 Eisenhower Avenue Alexandria, VA 22333	1
Director Ballistic Missile Defense Systems Command ATTN: BMDSC-TEN, Mr. N. J. Hurst BMDSC-LEI, Mr. J. Katechis BMDSC-LEI, Mrs. R. Buckelew BMDSC-AOLIB P.O. Box 1500 Huntsville, AL 35807	1 1 1 1
Ballistic Missile Defense Program Office ATTN: DACS-BMT DARCOM Bldg., Seventh Floor 5001 Eisenhower Avenue Alexandria, VA 22333	1
Director Ballistic Missile Defense Advanced Technology Center ATTN: ATC-X, Dr. J. Carlson ATC-X, Col. F. Gray ATC-R, D. Russ ATC-R, Maj. J. Hovious ATC-R, M. Whitfield ATC-M, D. Harmon ATC-M, J. Papadopoulos P.O. Box 1500 Huntsville, AL 35807	1 1 1 1 1 1 1



No. of Copies

Director  
Defense Nuclear Agency  
ATTN: SPAS, Dr. J. Slaughter 1  
Washington, DC 20305

Director  
Army Ballistic Research Laboratories  
ATTN: DRDAR-BLT, Dr. N. J. Huffington, Jr. 1  
DRDAR-BLT, Dr. T. W. Wright 1  
DRDAR-BLT, Dr. G. L. Moss 1  
Aberdeen Proving Ground, MD 21005

Commander  
Harry Diamond Laboratories  
ATTN: DRXDO-NP, Dr. F. Wimenitz 1  
2800 Powder Mill Road  
Adelphi, MD 20783

Commander  
Air Force Materials Laboratory  
Air Force Systems Command  
ATTN: LNE/Dr. W. Kessler 1  
LNC/Dr. D. Schmidt 1  
Wright-Patterson Air Force Base  
Dayton, OH 45433

Commander  
BMO/ABRES Office  
ATTN: BMO/MNRT, Col. R. Smith 1  
BMO/MNRTE, Maj. J. Sikra 1  
BMO/MNRTE, Maj. K. Yelmgren 1  
Norton Air Force Base, CA 92409

Commander  
Air Force Materials Laboratory  
ATTN: AFML/MBM, Dr. S. W. Tsai 1  
Wright-Patterson Air Force Base  
Dayton, OH 45433

Commander  
Naval Ordnance Systems Command  
ATTN: ORD-03331, Mr. M. Kinna 1  
Washington, DC 20360

No. of Copies

Commander	
Naval Surface Weapons Center	
ATTN: Dr. C. Lyons	1
Dr. W. Messick	1
Silver Springs, MD 20910	
Lawrence Livermore National Laboratory	
ATTN: Dr. E. M. Wu	1
P.O. Box 808 (L-342)	
Livermore, CA 94550	
Sandia Laboratories	
ATTN: Dr. Frank P. Gerstle, Jr.	1
Dr. L. D. Bertholf	1
Dr. J. Lipkin	1
P.O. Box 5800	
Albuquerque, NM 87115	
Aerospace Corporation	
ATTN: Dr. R. Cooper	1
P.O. Box 92975	
Los Angeles, CA 90009	
AVCO Corporation	
Government Products Group	
ATTN: Dr. W. Reinecke	1
Mr. P. Rolincik	1
201 Lowell Street	
Wilmington, MA 01997	
Boeing Commercial Airplane Company	
ATTN: Dr. K. Y. Lin	1
Mail Stop 6W-13	
Seattle, WA 98124	
ETA Corporation	
ATTN: D. L. Mykkanen	1
P.O. Box 6625	
Orange, CA 92667	
Effects Technology, Inc.	
ATTN: Dr. R. Wengler	1
Dr. R. Parris	1
J. Green	1
5383 Hollister Avenue	
Santa Barbara, CA 93111	

No. of Copies

Fiber Materials, Inc.  
ATTN: M. Subilia, Jr. 1  
L. Landers 1  
R. Burns 1  
Biddeford Industrial Park  
Biddeford, ME 04005

General Electric Company  
Advanced Materials Development Laboratory  
ATTN: Ms. B. Maguire, Room 5400B 1  
L. Pochettino, Room 5300B 1  
K. Hall, Room 5700 1  
J. Brazel, Room 5411B 1  
D. Stewart, Room 5211 1  
3198 Chestnut Street  
Philadelphia, PA 19101

General Dynamics Corporation  
Convair Division  
ATTN: J. Hertz 1  
H. McCutcheon, Jr. 1  
5001 Kearny Villa Road  
San Diego, CA 92138

General Dynamics Corporation  
ATTN: Dr. D. J. Wilkins, Mail Zone 2884 1  
P.O. Box 748  
Fort Worth, TX 76101

Kaman Sciences Corporation  
ATTN: D. Williams 1  
P.O. Box 7463  
Colorado Springs, CO 80933

Ktech  
ATTN: Dr. D. Keller 1  
911 Pennsylvania Avenue, N.E.  
Albuquerque, NM 87110

Lehigh University  
Institute of Fracture and Solid Mechanics  
ATTN: Dr. George C. Sih 1  
Bldg. 19, Packard Lab  
Bethlehem, PA 18015

Lockheed Missiles and Space Company  
ATTN: D. Aspinwall 1  
P.O. Box 504  
Sunnyvale, CA 94088

No. of Copies

Martin Marietta Aerospace	1
ATTN: V. Hewitt	1
Frank H. Koo	
P.O. Box 5837	
Orlando, FL 32805	
Massachusetts Institute of Technology	
Department of Aeronautics and Astronautics	
ATTN: Professor Theodore H. Pian	1
Professor James Mar	1
77 Massachusetts Avenue	
Cambridge, MA 02139	
McDonnell Douglas Corporation	
ATTN: Dr. L. Cohen	1
H. Parachanian	1
5301 Bolsa Avenue	
Huntington Beach, CA 92647	
Pacifica Technology, Inc.	
ATTN: Dr. Ponsford	1
P.O. Box 148	
Del Mar, CA 92014	
University of Illinois at Chicago	
Department of Civil Engineering,	
Mechanics and Metallurgy	1
ATTN: Professor R. L. Spilker	
Box 4548, Chicago, IL 60680	
Radkowski Associates	
ATTN: Dr. P. Radkowski	1
P.O. Box 5474	
Riverside, CA 92507	
Rensselaer Polytechnic Institute	
Department of Applied Mechanics	
ATTN: Professor E. H. Lee	1
Troy, NY 12181	
Southwest Research Institute	
ATTN: A. Wenzel	1
8500 Culebra Road	
San Antonio, TX 78206	
Terra Tek, Inc.	
ATTN: Dr. A. H. Jones	1
420 Wakara Way	
Salt Lake City, UT 84103	

No. of Copies

Defense Documentation Center 1  
Cameron Station, Bldg. 5  
5010 Duke Station  
Alexandria, VA 22314

Director

Army Materials and Mechanics Research Center  
ATTN: DRICMR-B, J. F. Dignam 1  
DRICMR-B, Dr. S. C. Chou 10  
DRICMR-B, L. R. Aronin 1  
DRICMR-B, Dr. D. P. Dandekar 1  
DRICMR-K, Procurement Office 1  
DRICMR-PL, Library 2  
DRICMR-PR, Technical Reports Office 1  
Watertown, MA 02172

AD Unclassified  
Unlimited Distribution

Army Materials and Mechanics Research Center  
Watertown, Massachusetts 02172  
SINGULARITIES AT THE TIP OF A CRACK NORMAL  
TO THE INTERFACE OF AN ANISOTROPIC LAYERED  
COMPOSITE  
T. C. T. Ting and P. H. Hoang  
Department of Civil Engineering, Mechanics  
and Metallurgy  
University of Illinois at Chicago  
P.O. Box 4348, Chicago, Illinois 60680  
Technical Report AMRC TR 83-28, May 1983, 42 pp  
illus.-tables, Contract DAAG46-81-K-0086  
D/A Project: 8H363304D215;  
AMCMS Code: 633000.21506  
Final Report, October 1981 to November 1982

Key Words  
Composite materials  
Stress concentration  
Complex variables  
Eigenvalues  
Anisotropy  
Elasticity

The order of stress singularities at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic layers in a composite is studied. Assuming that the stress singularities have the form  $r^{-k}$ , equations are derived for determining the order of singularities  $k$ . If the materials on both sides of the interface are identical,  $k = \frac{1}{2}$  is a root of multiplicity three each of which can be identified with the singularity due to, respectively, a symmetric tensile stress applied at infinity, an antisymmetric plane shear stress and an antiplane shear stress applied at infinity. When the materials on both sides of the interface are not the same, there are in general three distinct roots for  $k$ . Numerical examples for a typical high modulus graphite/epoxy and for a special T300/5208 graphite/epoxy show that  $k$  has three positive roots all of which are close to  $\frac{1}{2}$  for most combinations of ply-angles in the two materials.

AD Unclassified  
Unlimited Distribution

Army Materials and Mechanics Research Center  
Watertown, Massachusetts 02172  
SINGULARITIES AT THE TIP OF A CRACK NORMAL  
TO THE INTERFACE OF AN ANISOTROPIC LAYERED  
COMPOSITE  
T. C. T. Ting and P. H. Hoang  
Department of Civil Engineering, Mechanics  
and Metallurgy  
University of Illinois at Chicago  
P.O. Box 4348, Chicago, Illinois 60680  
Technical Report AMRC TR 83-28, May 1983, 42 pp  
illus.-tables, Contract DAAG46-81-K-0086  
D/A Project: 8H363304D215;  
AMCMS Code: 633000.21506  
Final Report, October 1981 to November 1982

Key Words  
Composite materials  
Stress concentration  
Complex variables  
Eigenvalues  
Anisotropy  
Elasticity

The order of stress singularities at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic layers in a composite is studied. Assuming that the stress singularities have the form  $r^{-k}$ , equations are derived for determining the order of singularities  $k$ . If the materials on both sides of the interface are identical,  $k = \frac{1}{2}$  is a root of multiplicity three each of which can be identified with the singularity due to, respectively, a symmetric tensile stress applied at infinity, an antisymmetric plane shear stress and an antiplane shear stress applied at infinity. When the materials on both sides of the interface are not the same, there are in general three distinct roots for  $k$ . Numerical examples for a typical high modulus graphite/epoxy and for a special T300/5208 graphite/epoxy show that  $k$  has three positive roots all of which are close to  $\frac{1}{2}$  for most combinations of ply-angles in the two materials.

AD Unclassified  
Unlimited Distribution

Army Materials and Mechanics Research Center  
Watertown, Massachusetts 02172  
SINGULARITIES AT THE TIP OF A CRACK NORMAL  
TO THE INTERFACE OF AN ANISOTROPIC LAYERED  
COMPOSITE  
T. C. T. Ting and P. H. Hoang  
Department of Civil Engineering, Mechanics  
and Metallurgy  
University of Illinois at Chicago  
P.O. Box 4348, Chicago, Illinois 60680  
Technical Report AMRC TR 83-28, May 1983, 42 pp  
illus.-tables, Contract DAAG46-81-K-0086  
D/A Project: 8H363304D215;  
AMCMS Code: 633000.21506  
Final Report, October 1981 to November 1982

Key Words  
Composite materials  
Stress concentration  
Complex variables  
Eigenvalues  
Anisotropy  
Elasticity

The order of stress singularities at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic layers in a composite is studied. Assuming that the stress singularities have the form  $r^{-k}$ , equations are derived for determining the order of singularities  $k$ . If the materials on both sides of the interface are identical,  $k = \frac{1}{2}$  is a root of multiplicity three each of which can be identified with the singularity due to, respectively, a symmetric tensile stress applied at infinity, an antisymmetric plane shear stress and an antiplane shear stress applied at infinity. When the materials on both sides of the interface are not the same, there are in general three distinct roots for  $k$ . Numerical examples for a typical high modulus graphite/epoxy and for a special T300/5208 graphite/epoxy show that  $k$  has three positive roots all of which are close to  $\frac{1}{2}$  for most combinations of ply-angles in the two materials.

AD Unclassified  
Unlimited Distribution

Army Materials and Mechanics Research Center  
Watertown, Massachusetts 02172  
SINGULARITIES AT THE TIP OF A CRACK NORMAL  
TO THE INTERFACE OF AN ANISOTROPIC LAYERED  
COMPOSITE  
T. C. T. Ting and P. H. Hoang  
Department of Civil Engineering, Mechanics  
and Metallurgy  
University of Illinois at Chicago  
P.O. Box 4348, Chicago, Illinois 60680  
Technical Report AMRC TR 83-28, May 1983, 42 pp  
illus.-tables, Contract DAAG46-81-K-0086  
D/A Project: 8H363304D215;  
AMCMS Code: 633000.21506  
Final Report, October 1981 to November 1982

Key Words  
Composite materials  
Stress concentration  
Complex variables  
Eigenvalues  
Anisotropy  
Elasticity

The order of stress singularities at the tip of a crack which is normal to and ends at an interface between two anisotropic elastic layers in a composite is studied. Assuming that the stress singularities have the form  $r^{-k}$ , equations are derived for determining the order of singularities  $k$ . If the materials on both sides of the interface are identical,  $k = \frac{1}{2}$  is a root of multiplicity three each of which can be identified with the singularity due to, respectively, a symmetric tensile stress applied at infinity, an antisymmetric plane shear stress and an antiplane shear stress applied at infinity. When the materials on both sides of the interface are not the same, there are in general three distinct roots for  $k$ . Numerical examples for a typical high modulus graphite/epoxy and for a special T300/5208 graphite/epoxy show that  $k$  has three positive roots all of which are close to  $\frac{1}{2}$  for most combinations of ply-angles in the two materials.

**END**

**FILMED**

**9-83**

**DTIC**