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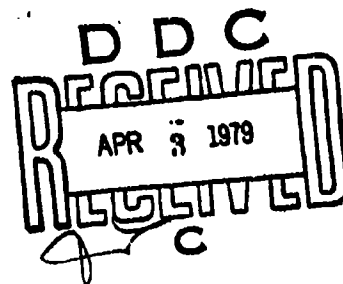
COMBUSTOR DESIGN CRITERIA VALIDATION
Volume III - User's Manual

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APPLIED TECHNOLOGY LABORATORY POSITION STATEMENT

This report describes an effort undertaken to improve small gas turbine combustor design techniques. This analytical procedure is viewed as a significant step toward reducing the design and development time and the cost associated with future Army gas turbine combustors while simultaneously achieving a more durable and fuel-efficient design. The reader is referred to the report documentation page for a description of each of the three volumes of this report. It is considered worthy of widespread application with the turbine industry. Any critique or other response regarding its use should be addressed to this Laboratory.

Mr. Kent Smith of the Propulsion Technical Area, Aeronautical Technology Division, served as Project Engineer for this effort.

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I. INTRODUCTION

GENERAL INFORMATION

The present program represents an extension and refinement of the previous effort with specific application to the design requirements of advanced, small, high-temperature-rise combustors for aircraft engines in the 2- to 5-pound-per-second (0.91- to 2.27-kilogram-per-second) flow range. This program was performed for the Applied Technology Laboratory, U.S. Army Research and Technology Laboratories (AVRADCOM), Ft. Eustis, Virginia by the AiResearch Manufacturing Company of Arizona during the period July, 1975, to October, 1978. The program is documented in this three-volume final report.

OBJECTIVE

The primary objective of this program was to further develop and validate existing analytical combustor design procedures that can be used to significantly shorten the design and development cycle of small advanced gas turbine engine combustors. Descriptions of the combustor analytical models, element tests and model validations are presented in Volume I.

The basic approach of the program consisted of a concentrated analytical treatment of key combustion phenomena affecting combustor performance complemented by rig tests. The rig test culminated in a complete series of performance mapping to validate the empirical/analytical combustor design procedure in an environment matching an actual operating engine.

The program was initially comprised of four technical tasks:

Task I - Analytical-Model Refinement

Task II - Full-Scale Combustor Design, Fabrication, and Preliminary Tests

Task III - Combustor-Performance Mapping

Task IV - Limited Modification and Retest

The Task I technical effort is described in Volume I. A complete description of the Task II and Task III activities is presented in Volume II. The computer codes for combustor design that evolved from that effort are fully documented in Volume III of this report. The combustor performance goals were achieved in Tasks II and III; thus Task IV was cancelled.

The computer models are based upon the numerical solution of the governing aero/thermo equations applicable to turbo-propulsion combustor environment, and are, therefore, applicable for analyzing internal flow field of can, can-annular and annulus combustor geometries. Both the inline and reverse-flow combustor configurations can be analyzed.

The cost-effectiveness of the empirical/analytical design procedure was to be demonstrated by undertaking the design and development testing of two full-scale annular combustors based on the following engine/combustor configurations, parameters and goals:

1. Engine/Component Configurations.

- Annular-combustor configurations
- Centrifugal compressor (last stage)
- First-stage axial turbine
- Nonregenerative cycle

2. Parameters and Goals.

- Engine airflow, $W_{a3} = 2.87$ pounds per second (1.30 kg/s)
- Combustor inlet pressure (P_3) = 10 atmospheres
- Compressor efficiency = 78.4 percent (total-to-static)
- Combustor inlet temperature = 660°F (622°K)
- Combustion efficiency = 99.5 percent (100 percent power) = 98.0 percent (5 percent power)
- Combustor pressure loss $\frac{P_{T3} - P_{T4}}{P_{T3}} = 3$ percent
- Combustor discharge temperature (T_{4avg}) = 2300°F (1533°K)
- Maximum pattern factor (PF) ≤ 0.23

where $PF = \frac{T_{4 \max} - T_{4avg}}{T_{4avg} - T_3}$

- Average radial temperature profile compatible with typical turbine blade requirements
- Maximum radial pattern factor (RPF) ≤ 0.075

where $PF = \frac{T_{4 \text{ avg rad max}} - T_{4 \text{ avg}}}{T_{4 \text{ avg}} - T_3}$

$T_{4 \text{ avg rad max}}$ = peak value of the circumferentially averaged radial temperature profile

- Good light-off/relight capability to 20,000 feet (6091 meters) altitude and ambient-temperature conditions per MIL-E-5007D Paragraph 3.2.5.1 (dated 15 October 1973)
- No visible carbon formation with hot fuel or at high-altitude conditions
- Multifuel capability, including JP-4 and JP-5
- Fuel contamination tolerance per MIL-E-8593A, Table X with filtration to 10 microns
- The combined CO and HC exhaust emissions will be sufficiently low to meet the previously noted combustion efficiency goals at 100- and 5-percent rated power. The NO_xLTO emissions level will be at or below the 1979 EPA NO_x standards. The maximum smoke number will be below the threshold of the exhaust plume visibility
- Acceptable component temperature levels and gradients to ensure long combustion system life and reliability
- Reasonable cost and weight

SUMMARY

A complete description of the following six combustor analytical models, associated computer codes, and users manuals are given in this report.

- Annulus-flow model
- Combustor-performance model

- Liner-cooling model
- Transition-liner mixing model
- Emissions model
- Fuel-insertion model

The annulus-flow model is used to compute airflow distribution around the combustor liner and pressure drop. The information provided by this model on jet velocities and efflux angles is used for specifying the boundary conditions of the internal-flow computer models.

A 3-D reacting recirculating-flow model is used for computing internal profiles of velocity components, chemical species, and temperature of a given combustor design. Effects of detail-design changes can be analytically predicted in regard to combustion efficiency, exhaust temperature quality, and lean blowout.

Two-dimensional parabolic programs are used for predicting liner-wall-temperature levels, mixing rate in the combustor transition liner of reverse-flow annular combustors, and gaseous emissions. A fuel-insertion model is used to compute mean drop-let size and size distribution of pressure atomizers, including simplex, duplex, and air-assist pressure atomizers, and airblast nozzles. The model is also used to compute spray trajectory and evaporation rate of a given nozzle design in a specified flow field.

The use of these models is illustrated in Section IV by a worked-out illustration of a simple combustor.

II. DESCRIPTION OF ANALYTICAL MODELS

The six combustor analytical models are described in this section. The relevant computer codes are described in Section III. The use of these models is illustrated by an example in Section IV.

ANNULUS-FLOW MODEL

An annulus-flow model is used to compute pressure losses, annulus Mach number and associated air velocity, and airflow distribution around the combustor liner.

A one-dimensional analysis of the plenum annulus is conducted based upon the generalized one-dimensional continuous flow-analysis approach of Shapiro¹. The analysis considers the effect of area change, wall friction, drag introduced by inserted obstacles such as fuel nozzles and service struts, heat transfer from the liner wall, and injection or extraction of air from the annulus. The analysis is valid for constant specific heat and molecular weight.

Following the approach of Shapiro for a small control volume around a point P located at a distance "X" from the compressor discharge, as shown in Figure 1, the following three working relations are obtained for Mach number M, stagnation pressure P_0 , and static temperature T.

¹Shapiro, Ascher H., "The Dynamics and Thermodynamics of Compressible Fluid Flow, Vol. I", Chapter B, The Ronald Press Company, New York (1953).

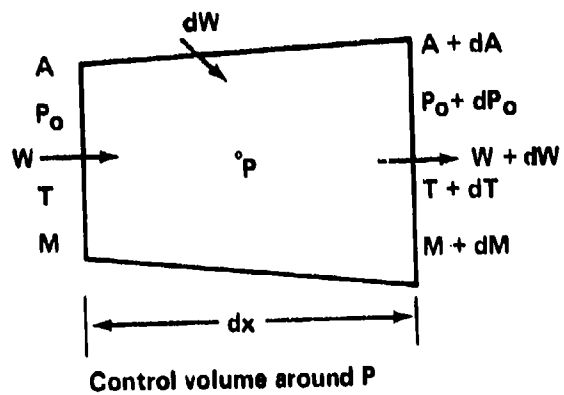
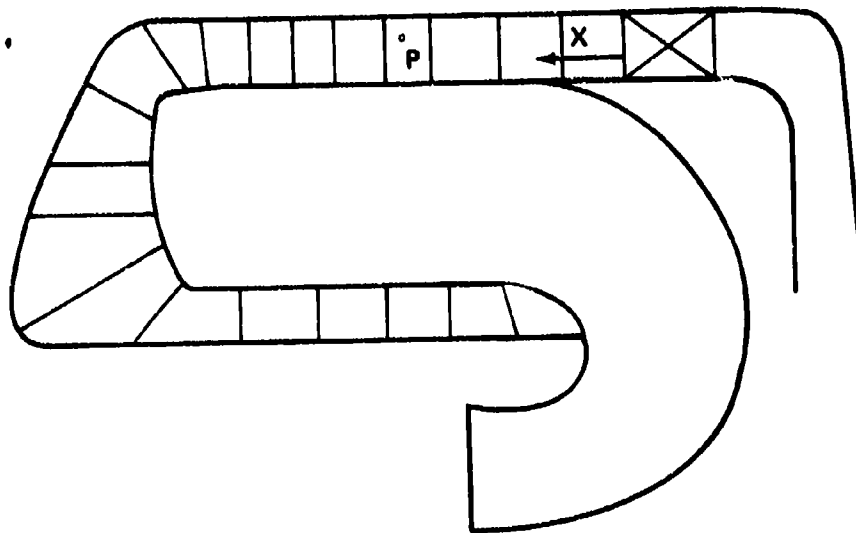


Figure 1. Control Volumes for Annulus-Flow Model.

$$\frac{dM^2}{M^2} = -2 \left(\frac{1+\gamma-1}{2} \frac{M^2}{1-M^2} \right) \left[\frac{dA}{A} + \left(\frac{1+\gamma M^2}{2} \right) \frac{dT_o}{T_o} + \frac{\gamma M^2}{2} \left(4f \frac{dx}{D_H} + \frac{dF}{\frac{1}{2}\gamma \rho A M^2} - 2\gamma \frac{dW}{W} \right) + (1+\gamma M^2) \frac{dW}{W} \right] \quad (1)$$

$$\frac{dp_o}{p_o} = -\frac{\gamma M^2}{2} \left[\frac{dT_o}{T_o} + 4f \frac{dx}{D_H} + \frac{dF}{\frac{1}{2}\gamma \rho A M^2} + 2(1-\gamma) \frac{dW}{W} \right] \quad (2)$$

$$\begin{aligned} \frac{dT}{T} &= \frac{M^2}{1-M^2} \left[(\gamma-1) \frac{dA}{A} + \left(\frac{1-\gamma M^2}{M^2} \right) \left(\frac{1+\gamma-1}{2} M^2 \right) \frac{dT_o}{T_o} \right. \\ &\quad - \frac{\gamma}{2} (\gamma-1) M^2 \left(4f \frac{dx}{D_H} + \frac{dF}{\frac{1}{2}\gamma \rho A M^2} - 2\gamma \frac{dW}{W} \right) \\ &\quad \left. - (\gamma-1) (1+\gamma M^2) \frac{dW}{W} \right] \quad (3) \end{aligned}$$

Where:

A = Flow area

D_H = Mean hydraulic diameter

f = Coefficient of skin friction

F = Drag force by inserted obstacles

T = Static gas temperature

W = Mass flow rate
y = Injected mass axial velocity/mainstream velocity
 γ = Ratio of specific heats

The above set of equations is written for each of the control volumes, shown schematically in Figure 1, applicable to a reverse-flow combustor geometry. Appropriate expressions are used for skin friction coefficient and drag introduced by fuel nozzles and other obstacles in the flow path. The remaining unknown variable dW , which will be negative for the flow through various orifices, is calculated by using the following approach.

The orifice configurations used in a combustor liner can be broadly divided into two basic categories.

- Configurations such as swirlers, primary pipes, and venturi sections which are either difficult to handle analytically or their flow rates are less affected by approach conditions.
- Liner orifices, including flush port, plunged holes, and scooped ports are affected by approach conditions and are amenable to analytical approach for predicting flow rates, jet velocities, and efflux angles.

The first type of ports are handled by specifying discharge coefficients, whereas the liner orifices are handled by using a modified analytical approach described by Gurevich². The Gurevich approach is based upon a 2-D potential flow solution of a problem shown schematically in Figure 2.

For an infinitely long slot of width b , the following three relations are obtained for the three unknowns, namely n , β and C_D .

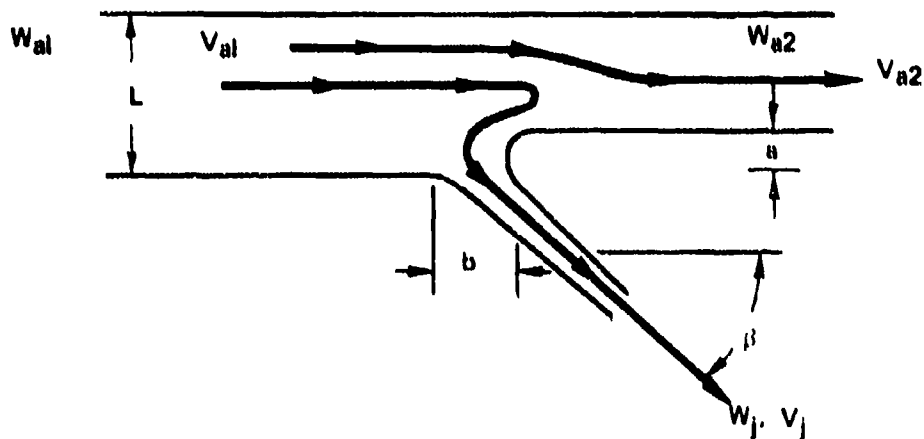
²Gurevich, M.I., "Theory of Jets in an Ideal Fluid", Pergamon Press, pp 52-59.

$$c = \cos\beta / (1 - \frac{1}{2n}) \quad (4)$$

$$\frac{b}{L} = \frac{1}{n} \left[\frac{\pi \sin\beta - \frac{\cos\beta}{n} \ln\left(\frac{1 + \cos\beta}{1 - \cos\beta}\right)}{\left(1 - \frac{1}{2n}\right)^2 + \cos^2\beta} + \frac{\ln\left\{\frac{\left(1 - \frac{1}{2n}\right) + \cos\beta}{\left(1 - \frac{1}{2n}\right) - \cos\beta}\right\}}{\left(1 - \frac{1}{2n}\right) \cos\beta} \right] \quad (5)$$

$$\frac{\left(1 - \frac{1}{2n}\right)^2 + \left(1 - \frac{1}{n}\right)^2 \cos^2\beta}{\left(1 - \frac{1}{2n}\right) \cos\beta} \ln\left\{\frac{\left(1 - \frac{1}{2n}\right) + \left(1 - \frac{1}{n}\right) \cos\beta}{\left(1 - \frac{1}{2n}\right) - \left(1 - \frac{1}{n}\right) \cos\beta}\right\} \left[\frac{\cos\beta}{\left(1 - \frac{1}{2n}\right)}\right]$$

$$C_D = \frac{1}{b} \left(\frac{2 \cos\beta}{2n - 1} \right) \quad (6)$$



Where:

- a/L = annulus width change/upstream width
- b/L = slot width/upstream annulus width
- n = annulus upstream flow rate/slot flow rate, W_{al}/W_j
- c = upstream annulus velocity/slot velocity, V_{al}/V_j
- β = jet efflux angle
- C_D = discharge coefficient

Figure 2. Liner Port Model Schematic.

These equations are applied to combustor liner orifices by maintaining area similarity through the following relationships:

$$b/L = A_H/A_{ea}$$

where A_H = orifice area ($= \frac{\pi}{4} D^2$ for circular hole)

A_{ea} = effective annulus area with boundary-layer blockage effects

For a given application, the annulus upstream conditions and the static pressure inside the combustor must be specified. With the above equations, an orifice can be sized to pass a specified flow rate, or the flow through a specified orifice can be calculated. The procedure for each is outlined as follows.

For given values of annulus and orifice flow rates and velocities, c can be calculated and then the efflux angle can be found from Equation 4. For the special case where all annulus flow passes through the orifice, $n = 1$ and

$$\cos \beta = c/2$$

If the orifice flow is a negligible portion of the annulus flow, n approaches infinity,

$$\cos \beta = c$$

After the value of β is obtained, Equation 5 can be used to calculate b/L and A_H/A_{ea} from Equation 7; then from Equation 6, C_D can be calculated.

For the alternate problem, with the orifice specified, the above procedure is used in an iterative solution starting with an estimated flow rate (value of n). The iteration is continued until n converges to a small difference between iterations.

Such an approach has given good correlation with measure C_D^3 data of the circular orifices, as shown typically in Figure 3. For plunged orifices and the metering orifices of film-cooling geometries, the approach gave only qualitative agreement. However, by multiplying the computed C_D values by 1.48 and 1.4, and by assuming β equal to 80- and 0-degrees, respectively, for the plunged orifices and the cooling slot, the approach gave good agreement with the data, as shown in Figure 3.

With the above procedure for computing dW appearing in Equations 1, 2, and 3, it is now possible to write a set of equations for each of the control volumes around the combustor liner, as shown schematically in Figure 1. These equations are solved iteratively to compute isothermal combustor-pressure drop. To this can be added pressure drop due to heat addition which gives combustor total pressure drop.

3-D COMBUSTOR PERFORMANCE MODEL

A 3-D elliptic, reacting-flow model is used for calculating internal flow field of gas turbine combustors. The model solves governing equations for the following variables, as described in Paragraph 6 below:

- Axial, radial, and swirl velocity components
- Specific enthalpy and temperature
- Turbulence kinetic energy and dissipation rate

³Hunter, S. C., K. M. Johansen, H. C. Mongia, and M. P. Wood, "Advanced, Small, High-Temperature-Rise Combustor Program, Volume II: Analytical Mode Derivation and Combustor-Element Rig Tests (Phases I and II)", AD778766 (1974).

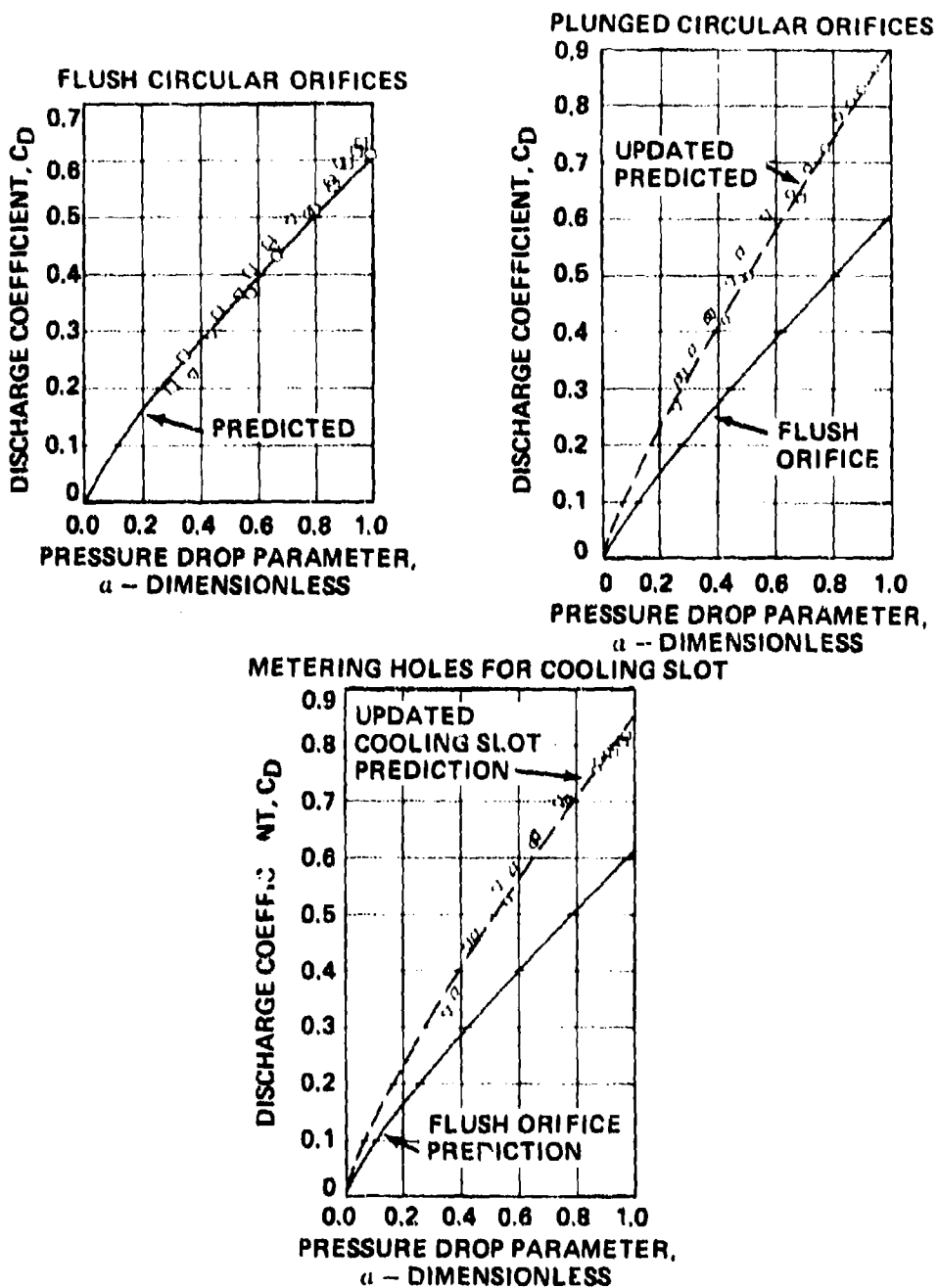


Figure 3. A Typical Comparison Between Prediction and Measured C_D .

- Unburned fuel, CO, and composite fuel fraction
- Radiation-flux vectors
- Spray combustion

Paragraphs 7 and 8 below give a brief description of the boundary conditions and the numerical scheme used for solving the set of nonlinear coupled partial differential equations.

1. Equations of Continuity, Momentum, and Enthalpy.

a. Continuity.

$$\text{div} (\rho \vec{V}) = \dot{m}'_{\text{spray}} \quad (8)$$

b. x-Momentum.

$$\begin{aligned} \text{div} (\rho \vec{V} u - \mu_{\text{eff}} \text{grad } u) = & - \frac{\partial p}{\partial x} - \frac{2}{3} \frac{\partial}{\partial x} (\mu_{\text{eff}} \text{div } \vec{V}) \\ & + \frac{\partial}{\partial x} (\mu_{\text{eff}} \frac{\partial u}{\partial x}) + \frac{1}{r} \frac{\partial}{\partial r} (\mu_{\text{eff}} r \frac{\partial v}{\partial x}) + \frac{1}{r} \frac{\partial}{\partial \theta} (\mu_{\text{eff}} \frac{\partial w}{\partial x}) \\ & + s_{\text{spray}}^u \end{aligned} \quad (9)$$

c. y-Momentum.

$$\begin{aligned} \text{div} (\rho \vec{V} v - \mu_{\text{eff}} \text{grad } v) = & - \frac{\partial p}{\partial r} - \frac{2}{3} \frac{\partial}{\partial r} (\mu_{\text{eff}} \text{div } \vec{V}) \\ & + \frac{\partial}{\partial x} (\mu_{\text{eff}} \frac{\partial u}{\partial r}) + \frac{1}{r} \frac{\partial}{\partial r} (\mu_{\text{eff}} r \frac{\partial v}{\partial r}) + \frac{1}{r} \frac{\partial}{\partial \theta} [\mu_{\text{eff}} (\frac{\partial w}{\partial r} - \frac{w}{r})] \\ & - 2 \frac{\mu_e}{r} (\frac{1}{r} \frac{\partial w}{\partial \theta} + \frac{v}{r}) + \frac{\rho w^2}{r} + s_{\text{spray}}^v \end{aligned} \quad (10)$$

d. θ -Momentum.

$$\begin{aligned} \text{div} (\rho \vec{V} w - \mu_{\text{eff}} \text{grad } w) &= -\frac{1}{r} \frac{\partial p}{\partial \theta} - \frac{2}{3} r \frac{\partial}{\partial \theta} (\mu_{\text{eff}} \text{div } \vec{V}) \\ &+ \frac{\partial}{\partial x} \left(\frac{\mu_{\text{eff}}}{r} \frac{\partial u}{\partial \theta} \right) + \frac{1}{r} \frac{\partial}{\partial r} \left[\mu_{\text{eff}} r \left(\frac{1}{r} \frac{\partial v}{\partial \theta} - \frac{w}{r} \right) \right] \\ &+ \frac{1}{r} \frac{\partial}{\partial \theta} \left[\frac{\mu_{\text{eff}}}{r} \left(\frac{\partial w}{\partial \theta} + 2v \right) \right] - \frac{\rho v w}{r} + \frac{\mu_{\text{eff}}}{r} \left(\frac{\partial w}{\partial r} + \frac{\partial v}{r \partial \theta} - \frac{w}{r} \right) \\ &+ S_{\text{spray}}^w \end{aligned} \quad (11)$$

e. Specific enthalpy.

$$\text{div} \left(\rho \vec{V} h - \frac{\mu_{\text{eff}}}{P r_{\text{eff}}} \text{grad } h \right) = S_h \quad (12)$$

where

S_h represents the sum of all the enthalpy source terms for radiation and spray evaporation

Definition of variables are:

\vec{V} = Net gas velocity vector

u, v, w = Velocity components along x , radial and circumferential directions

x, y, θ = Axial, radial and circumferential coordinates

p = Static pressure

h = Static specific enthalpy

$S_{\text{spray}}^u, S_{\text{spray}}^v, S_{\text{spray}}^w$ = Momentum transfer from spray to the gas phase u, v and w - momentum equations

\dot{m}'_{spray} = spray evaporation/combustion rate per unit volume

$$\text{div} (\rho \vec{v} \phi) = \frac{1}{r} \left[\frac{\partial}{\partial x} (r \rho u \phi) + \frac{\partial}{\partial r} (r \rho v \phi) + \frac{\partial}{\partial \theta} (r \rho w \phi) \right]$$

$$\text{div} (\mu \text{grad } \phi) = \frac{1}{r} \left[\frac{\partial}{\partial x} (r \mu \frac{\partial \phi}{\partial x}) + \frac{\partial}{\partial r} (r \mu \frac{\partial \phi}{\partial r}) + \frac{\partial}{\partial \theta} (\frac{\mu}{r} \frac{\partial \phi}{\partial \theta}) \right]$$

The effective viscosity μ_{eff} is given by $\mu_{\text{eff}} = \mu_l + \mu_t$

where μ_l and μ_t are the molecular and turbulent viscosities of the fluid, respectively.

2. Turbulence Model.

The turbulent viscosity μ_t is calculated by using a two-equation turbulence model that solves governing equations for the turbulence kinetic energy (k) and the dissipation rate (ϵ). The governing equations for k and ϵ are:

$$\text{div} (\rho \vec{v} k - \Gamma_{k,\text{eff}} \text{grad } k) = G_k - \rho \epsilon \quad (13)$$

$$\text{div} (\rho \vec{v} \epsilon - \Gamma_{\epsilon,\text{eff}} \text{grad } \epsilon) = (C_1 G_k - C_2 \rho \epsilon) \frac{\epsilon}{k} \quad (14)$$

where

$$G_k = \mu_t \left[2 \left\{ \left(\frac{\partial u}{\partial x} \right)^2 + \left(\frac{\partial v}{\partial r} \right)^2 + \left(\frac{\partial w}{r \partial \theta} + \frac{v}{r} \right)^2 \right\} + \left(\frac{\partial w}{\partial x} + \frac{\partial u}{r \partial \theta} \right)^2 + \left(\frac{\partial u}{\partial r} + \frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial r} + \frac{\partial v}{r \partial \theta} - \frac{w}{r} \right)^2 \right] \quad (15)$$

$$\Gamma_{k,\text{eff}} = \mu_{\text{eff}} / \sigma_{k,\text{eff}} \quad (16)$$

$$\Gamma_{\epsilon,\text{eff}} = \mu_{\text{eff}} / \sigma_{\epsilon,\text{eff}}$$

$$\mu_t = C_D \rho k^2 / \epsilon$$

C_D , C_1 , and C_2 are constants. $\Gamma_{k,eff}$, $\Gamma_{\epsilon,eff}$, $\sigma_{k,eff}$ and $\sigma_{\epsilon,eff}$ are the effective exchange coefficients and Schmidt numbers for k and ϵ , respectively.

The k - ϵ turbulence model is moderate in complexity and is considered to be superior to other models having a similar degree of complexity. This model has been extensively used by many investigators and has proved to be adequate in a wide range of flow conditions. More advanced turbulence models, such as those based upon the Reynolds-stress modeling approach, are not yet fully developed to warrant their use in recirculating flow-field problems as encountered in gas-turbine combustors. In addition, such an approach will appreciably increase the computation effort.

Recommended values for the constants appearing in the above equations are

$$C_D = 0.09$$

$$C_1 = 1.44$$

$$C_2 = 1.92$$

$$\sigma_{k,eff} = 0.9$$

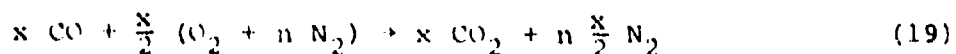
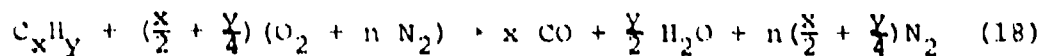
$\sigma_{\epsilon,eff}$ is calculated from

$$\sigma_{\epsilon,eff} = \frac{k^2}{(C_2 - C_1) C_D^{1/2}}$$

where k is the vonKarman constant taken to be equal to 0.42.

3. Chemical Species Equations.

A two-step kinetic scheme is used as represented by Equations 18 and 19.

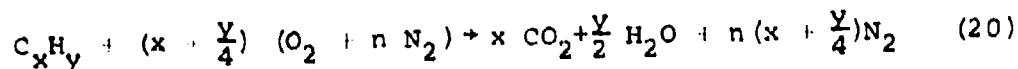


Stoichiometric oxygen-to-fuel and oxygen-to-CO ratios for the first and the second reactions are given by

$$i_1 = \frac{32 \left(\frac{x}{2} + \frac{y}{4} \right)}{(12x + y)}$$

$$i_2 = 0.571$$

Combining Equations 18 and 19, one obtains overall stoichiometry equation as



The corresponding stoichiometric oxygen-to-fuel ratio is

$$i = \frac{32 \left(x + \frac{y}{4}\right)}{(12x + y)}$$

Governing equations for fuel and CO mass fractions are:

$$\text{div} (\rho \vec{v} m_{fu} - \Gamma_{fu,eff} \text{grad} m_{fu}) = \dot{m}_{evap}''' - R_{fu} \quad (21)$$

$$\text{div} (\rho \vec{v} m_{CO} - \Gamma_{CO,eff} \text{grad} m_{CO}) = - R_{CO} \quad (22)$$

where R_{fu} and R_{CO} are rate of oxidation of fuel and CO in accordance with the combustion model explained in paragraph a. \dot{m}_{evap}''' is the rate of spray evaporation per unit volume computed in accordance with the spray combustion model description in Paragraph 5.

Equations similar to the above are required for O_2 , CO_2 , and H_2O . However, by using Shvab-Zeldovich approximation⁴, one needs to solve only one more equation for composite fuel fraction.

ϕ_{fuox}

⁴Williams, F. A., "Combustion Theory", Addison-Wesley Publishing Company, Inc. (1965).

$$\text{div} (\rho \vec{v} \phi_{\text{fuox}} - l'_{\text{fuox}} \phi_{\text{fuox}}) = \dot{m}'_{\text{evap}} \quad (23)$$

$$\text{where } \phi_{\text{fuox}} = \frac{\phi - \phi_A}{\phi_F - \phi_A}$$

$$\text{where } \phi = m_{\text{fu}} - \frac{m_{\text{ox}}}{i}$$

ϕ_A and ϕ_F are the values of ϕ for air and fuel streams, respectively.

Knowing the amount of fuel burned ($\text{FB} = \phi_{\text{fuox}} - m_{\text{fu}}$), the concentrations of CO_2 , O_2 , and H_2O are given by

$$m_{\text{CO}_2} = 44 \frac{x \text{ FB}}{(12 + y)} - \frac{44}{28} m_{\text{CO}}$$

$$m_{\text{ox}} = i m_{\text{fu}} + i_2 m_{\text{CO}} + 0.232 - (0.232 + i) \phi_{\text{fuox}}$$

$$m_{\text{H}_2\text{O}} = \frac{18 y \text{ FB}}{2 (12 + y)}$$

Mass fraction of N_2 is given by

$$m_{\text{N}_2} = 1 - (m_{\text{ox}} + m_{\text{fu}} + m_{\text{CO}} + m_{\text{CO}_2} + m_{\text{H}_2\text{O}})$$

a. Calculation of Reaction Rates.

Equations are needed for oxidation rates of fuel and CO , i.e., R_{fu} and R_{CO} . The turbulent reactive flow is an area of intensive research, and a number of models have been proposed to predict burning rate of fuel in turbulent environments. A simple model is used in the present study as explained in the following paragraphs.

The rate of oxidation of fuel is determined by the minimum of the following three equations:

$$R_{fu, ch} = k_1 \rho^{1.5} m_{ox} m_{fu}^{\frac{1}{2}} e^{-\left(\frac{E_1}{R_T}\right)} \quad (24a)$$

$$R_{fu, turb} = C_{R_1} \rho m_{fu} \epsilon/k \quad (24b)$$

$$R_{fuox, turb} = C_{R_1} \rho \frac{m_{ox}}{i_1} \epsilon/k \quad (24c)$$

Here, Equation 24a is the rate of fuel oxidation as controlled by chemical kinetics. Generally, a bimolecular Arrhenous expression is assumed for this reaction. However, the Task I reacting-flow mapping data was best correlated by using Equation 24a in conjunction with Equations 24b and 24c.

Equation 24b is based upon the eddy-breakup model of Spalding⁵ and expresses the rate of fuel oxidation as influenced by turbulence intensity and scale, and concentration of unburned fuel. This model is applicable to premixed flames. Since combustion in gas-turbine combustors is neither fully premixed nor entirely diffusion controlled, Equation 24c is postulated, similar to Equation 24, which determines the rate of fuel oxidation as controlled by the availability of the oxygen. The constant i_1 appears from stoichiometry of the chemical Equation 18. One could use i instead of i_1 without any loss of generality, because the empirical constant C_{R_1} will simply be different.

The rate of oxidation of CO is similarly the minimum of the following three equations:

$$R_{CO, ch} = k_2 \rho^2 m_{CO} m_{ox} e^{-\left(\frac{E_2}{R_T}\right)} \quad (25a)$$

$$R_{CO, turb} = C_{R_2} \rho m_{CO} \epsilon/k \quad (25b)$$

$$R_{CO, ox, turb} = C_{R_2} \rho \frac{m_{ox}}{i_2} \epsilon/k \quad (25c)$$

⁵Spalding, D. B., "Mixing and Chemical Reaction in Steady Confined Turbulent Flames", Thirteenth Symposium (International) on Combustion, The Combustion Institute, 1971.

4. Radiation Model.

A six-flux radiation model based upon the Schuster-Hamaker approximation^{6,7} is used in the present program. It should be noted that, as pointed out by Siddall,⁸ other flux model approximations such as Milne-Eddington and Schuster-Schwarzschild can be represented by the same form of flux equations with constants being different. Therefore, the user can modify the flux equations with relative ease.

The differential equations describing the variations of the fluxes along the six directions are:

$$\frac{d}{dx} (I_{x+}) = - (a + s) I_{x+} + aE + \frac{s}{6} I \quad (26)$$

$$\frac{d}{dx} (I_{x-}) = (a + s) I_{x-} - aE - \frac{s}{6} I \quad (27)$$

$$\frac{1}{r} \frac{d}{dr} (r I_{r+}) = - (a + s) I_{r+} + \frac{I_{r-}}{r} + aE + \frac{s}{6} I \quad (28)$$

$$\frac{1}{r} \frac{d}{dr} (r I_{r-}) = (a + s) I_{r-} + \frac{I_{r+}}{r} - aE - \frac{s}{6} I \quad (29)$$

$$\frac{1}{r} \frac{d}{d\theta} (I_{\theta+}) = - (a + s) I_{\theta+} + aE + \frac{s}{6} I \quad (30)$$

$$\frac{1}{r} \frac{d}{d\theta} (I_{\theta-}) = (a + s) I_{\theta-} - aE - \frac{s}{6} I \quad (31)$$

⁶Hamaker, H. C., "Radiation and Heat Conduction in Light-Scattering Material", Philips Research Report, Vol. 2, pp 55-67, 1947.

⁷Patankar, S. V., and D. B. Spalding, "A Computer Model for Three-Dimensional Flow in Furnaces", Fourteenth Symposium (International) on Combustion, The Combustion Institute, 1973.

⁸Siddall, R. G., "Flux Methods for the Analysis of Radiant Heat Transfer", Paper presented at the Fourth Symposium on Flames and Industry, 1972.

where I_{x+} , I_{r+} , and $I_{\theta+}$ are the fluxes along the positive directions of axial, radial and circumferential directions, respectively; I_{x-} , I_{r-} , and $I_{\theta-}$ are the corresponding fluxes along the negative directions.

a = absorption coefficient defined as radiation absorbed per unit length

s = scattering coefficient defined as radiation scattered per unit length

E = black body emissive power = σT^4

where σ is the Stefan-Boltzman constant

$$I = I_{x+} + I_{x-} + I_{r+} + I_{r-} + I_{\theta+} + I_{\theta-}$$

With the composite-fluxes R^x , R^r and R^z defined as:

$$R^x = \frac{1}{2} (I_{x+} + I_{x-})$$

$$R^r = \frac{1}{2} (I_{r+} + I_{r-})$$

$$R^z = \frac{1}{2} (I_{\theta+} + I_{\theta-})$$

one can reduce the six first-order flux equations into the following three second-order equations:

$$\frac{d}{dx} \left(\frac{1}{a+s} \frac{dR^x}{dx} \right) = a (R^x - E) + \frac{S}{3} (2R^x - R^r - R^z) \quad (32)$$

$$\frac{1}{r} \frac{d}{dr} \left(\frac{r}{a+s} \frac{dR^r}{dr} \right) = a (R^r - E) + \frac{S}{3} (2R^r - R^x - R^z) \quad (33)$$

$$\frac{1}{r} \frac{d}{d\theta} \left(\frac{1}{a+s} \frac{dR^z}{d\theta} \right) = a (R^z - E) + \frac{S}{3} (2R^z - R^x - R^r) \quad (34)$$

Once R^x , R^r , and R^z are known, the net radiation fluxes in the axial, radial, and circumferential directions, Q^x , Q^r , and Q^z respectively, are given by:

$$\begin{aligned} Q^x &= I_{x+} - I_{x-} \\ &= - \frac{2}{a+s} \frac{dR^x}{dx} \end{aligned} \quad (35)$$

$$\begin{aligned} Q^r &= I_{r+} - I_{r-} \\ &= - \frac{2}{a + s + \frac{1}{r}} \frac{dR^r}{dr} \end{aligned} \quad (36)$$

$$\begin{aligned} Q^z &= I_{\theta+} - I_{\theta-} \\ &= \frac{2}{a+s} \frac{1}{r} \frac{dR^z}{d\theta} \end{aligned} \quad (37)$$

The contribution of R^x , R^r , and R^z to the source terms of specific enthalpy, Equation 12, is given by:

$$\begin{aligned} (S_h)_{\text{radiation}} &= 2a [(R^x - E) + \\ &\quad (R^r - E) + \\ &\quad (R^z - E)] \end{aligned} \quad (38)$$

Since information on the variations of a and s (with other quantities such as concentrations of CO , H_2O , CO and soot particles) is often scarce and unprecise, they have been assumed to be uniform. However, variable values of a and s can be incorporated in the program with minor modifications.

5. Spray Combustion.

It is very important to predict aerodynamic interaction between evaporating/burning sprays and flow field insofar as combustion efficiency, pattern factor, stability, liner-wall temperature levels and gradients, smoke and gaseous emissions formation are concerned. For example, the presence of smaller droplets influence flame stabilization as they provide the main source of heat in the recirculation zone. Larger droplets, however, escape the recirculation zone and are mainly responsible for the smoke formation. Measured data by McCreath and Chigier⁹ showed that droplets with initial sizes less than 50 microns were evaporated in the recirculation zone. The smaller droplets were influenced greatly by the recirculation zone velocity field, whereas up to 70 percent of the bigger droplets in the 100-200 microns range escaped the recirculation zone. Their trajectory was not influenced by the flow-field velocity distribution. The flow-field influence on evaporation and trajectory of medium-size droplets, between 50 and 100 microns, was moderate.

Combustion characteristics of liquid droplets burning individually are significantly different from those burning collectively in a spray. For example, Beer and his associates¹⁰ showed that the burning-rate constant of monosized droplet arrays was about half that of single droplets. There was also significant reduction^{10,11} in drag coefficient, C_D , as compared to that of a

⁹McCreath, C. G. and N. A. Chigier, "Liquid Spray Burning in the Wake of a Stabilizer Disc", Fourteenth Symposium (International) on Combustion, The Combustion Institute (1973).

¹⁰Nuruzzaman, A. S. M., A. B. Hedley, and J. M. Beer, "Combustion of Monosized Droplet Streams in Self-Supporting Flames", Thirteenth Symposium (International) on Combustion, The Combustion Institute (1971).

¹¹Chigier, N. A., et. al, "Dynamics of Droplets in Burning and Isothermal Sprays", Combustion and Flame, V23 (1974).

nonreactive sphere. On the other hand, Natarajan¹² showed that C_D of a burning droplet should be calculated for a nonreacting sphere at the mean properties and initial diameter.

The quasi-steady droplet combustion theory with spherical symmetry predicts the burning rate constant K to be independent of the surrounding gas pressure, where $K = -\frac{d}{dt} d_L^2$, and d_L is the liquid-droplet diameter. However, it has been found experimentally¹³ that K increases with an increase in pressure. Raghunandan and Mukunda¹⁴ critically evaluated the quasi-steady approximation, variable gas phase properties and incomplete combustion as related to predictions of burning-rate constant, the flame-to-diameter ratio and the flame temperature. The liquid-phase unsteadiness lasts for about 20-25 percent of the total burning time. It was shown by the authors that a good correlation with the burning rate data could be obtained by taking thermal conductivity and C_D as a function of concentration and temperature.

A majority of the reported work has been concerned with combustion of single component hydrocarbon fuels. Limited work has been reported, such as Reference 15, for combustion of multi-component fuel droplets; but, these approaches became quite complicated for predicting spray combustion of complex fuels like jet aviation fuels.

¹²Natarajan, R., "Experimental Drag Coefficients for Evaporating and Burning Drops at Elevated Pressures", Combustion and Flame, V20 (1973).

¹³Rush, J. H., and H. Krier, "Burning of Fuel Droplets at Pressures Greater than Atmospheric", Combustion and Flame V22 (1974).

¹⁴Raghunandan, B. N., and H. S. Mukunda, "The Problem of Liquid Droplet Combustion - Reexamination", Combustion and Flame V30 (1977).

¹⁵Shyu, R. R., C. S. Chen, G. O. Gondie and M. M. Elwakil, "Multi-Component Heavy Fuel Drop Histories in a High Temperature Flow Field", Fuel, V51 (1972).

Figure 4 pictorially presents the approach that has been used for spray combustion in the present program. The spray cone is divided into a number of sections or rays. Each ray has a particular x, y, and z direction associated with it, depending on the orientation of the fuel nozzle in the combustor and the spray cone angle.

The initial conditions for each droplet are that they have a velocity as specified by the program user and a direction corresponding to the particular ray in question. The total fuel-flow rate is currently divided equally among the rays, although it would be easily possible to do otherwise. For each ray and for each droplet size group, of which five are assumed, the droplet trajectories are calculated from a force balance assuming the drag on the droplet is for that of a sphere. Heat transfer to the droplet is calculated using the coefficient given in Equation 39.

$$h = 2 \frac{k}{D} (1 + 0.3 \text{Pr}^{\frac{1}{3}} \text{Re}^{\frac{1}{2}}) \left(\frac{J}{m^2 - K} \right) \quad (39)$$

where k is the thermal conductivity of fuel vapor, Re is Reynolds number based on relative velocity, and D is the droplet diameter. Until the droplet reaches the boiling temperature, no evaporation is assumed to occur; however, once reached, the evaporation rate is obtained from the burning rate constant k . Where:

$$k_o = \frac{d}{dt} (D^2)$$

$$k_o = \frac{8}{\rho_f} \frac{1}{C_{P1}} \ln(1+B)$$

$$B = \text{mass transfer No.} = \frac{1}{L_{\text{vap}}} \left[m_{\text{O}_2} \frac{H_c}{i} + C_{P1} (T_{\infty} - T_{\beta}) \right]$$

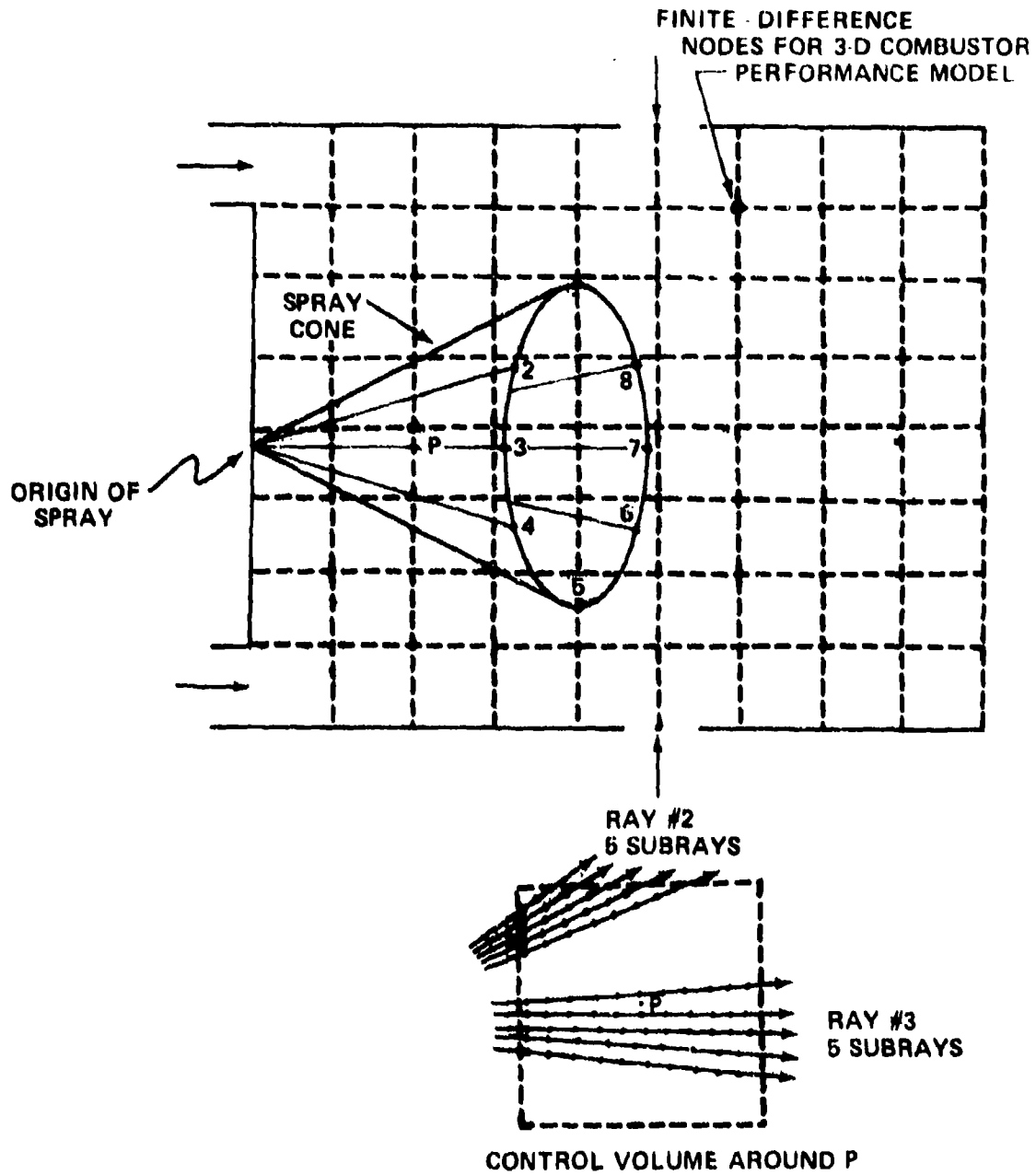


Figure 4. Schematic of Spray in Combustor Flow Field.

H_c = Heat of combustion

i = Stoichiometric O_2 -fuel mass ratio

L_{vap} = Latent heat of vaporization

ρ_f = Liquid fuel density

λ_1 and (CP_1) the thermal conductivity and specific heat inside the flame zone, are assumed to be

$$\lambda_1 = 0.4\lambda_f + 0.6 \lambda_{AIR}$$

$$CP_1 = CP_f$$

and are evaluated at the average of the boiling and flame temperatures. These calculations are performed explicitly, with care taken that the time step is sufficiently small, until at least 99 percent of the fuel has evaporated.

The above procedure requires knowledge of the fuel and air properties. The particular values used in the current program are listed below.

a. Droplet-Size Distribution.

<u>Group</u>	<u>Vol. % of Spray</u>	<u>Size Ratio (D/SMD)</u>
1	0-20	0.6
2	20-40	0.9
3	40-60	1.2
4	60-80	1.5
5	80-100	2.1

b. Liquid-Fuel Density (ρ_f).

$$\rho_f = 1000 [PR_{60} + 0.208 - 0.00072 T_f] \frac{KG}{m^3}$$

$$PR_{60} = \frac{1.076}{1 + \frac{1.076}{0.775} - 1 (1 - 0.67F)} \text{ for JP4}$$

PR_{60} = Specific gravity of residue at 60°F

F = Fraction evaporated

T_f = Fuel temperature

c. Specific Heat of Liquid Fuel (CP_f).

$$CP_f = 840.5 + 4.1372 T_f \text{ (J/KG } \cdot \text{ } ^\circ\text{K)}$$

d. Molecular Weight of Fuel Vapor.

Interpolated from table below,

<u>F</u>	<u>MW(JP4)</u>
0	93.26
0.1	114.60
0.3	126.61
0.5	138.16
0.7	150.59
0.9	173.21
1.0	204.76

e. Thermal Conductivity of Fuel Vapor (λ_f).

$$\lambda_f = 1.729 (A + BT_f) \text{ (j/M-}^\circ\text{K-Sec)}$$

where A and B are from the following list:

<u>MW_{fuel}</u>	<u>A</u>	<u>B</u>
50	-6.362E-3	53.5E-6
100	-6.358E-3	49.1E-6
150	-6.284E-3	46.6E-6
300	-6.010E-3	42.3E-6

f. Specific Heat of Fuel Vapor (CP_{f_v}).

$$CP_{f_v} = 4183.3 (0.153 + 0.00081T_f) \left(\frac{J}{KG-^{\circ}K}\right)$$

g. Latent Heat of Vaporization (L_{vap}).

$$L_{vap} = 30676.6 (1092.88 - 1.8T_f)^{.39} \quad (J/KG)$$

h. Boiling Temperature of Fuel (T_B).

$$T_B = A \ln P_v + B \quad (^{\circ}K)$$

where P_v is vapor pressure in pascals, A and B are from the following list:

<u>% Evaporated</u>	<u>A</u>	<u>B</u>
0	41.026	-114.000
10	30.857	41.574
30	27.348	96.534
50	23.997	146.567

The spray calculation procedure is briefly described in the following paragraph. Referring to Figure 4, the analysis is done for each of the rays selected and their subrays. Details are given for the control volume around a point P, where it is shown for two typical rays identified as Ray No. 2 and Ray No. 3. With each of these rays there might be five subrays, or less, depending upon the location of point P, initial droplet sizes, and the properties of the field through which the individual drops have traveled. Depending upon the direction of the ray and the

finite-difference nodal volume, calculations for evaporation/burning are done for a number of subgrid points. The droplets are allowed to exchange mass, momentum, and energy with the surrounding gas phase. The net amount of mass, energy, and momentum received by the node P is the sum total of all droplets passing through the control volume of P.

6. Calculation of Gas Temperature.

With the specific enthalpy and chemical species known, the gas temperature is calculated as follows: The specific enthalpy h is the summation of the enthalpies of individual species, i.e.,

$$\begin{aligned}
 h &= \sum_i m_i h_i \\
 &= \sum_i m_i \left[h_{i,0} + \int_{T_0}^T c_{pi}(T) dT \right] \\
 &= \sum_i m_i \left[h_{i,0} + \int_{T_0}^{T^*} c_{pi}(T) dT + \int_{T^*}^T c_{pi}(T) dT \right]
 \end{aligned}$$

Thus, giving

$$h = \sum_i m_i [h_i(T^*) + c_{pi}(T^*)(T-T^*)] \quad (47)$$

where m_i , h_i , $h_{i,0}$, c_{pi} , T^* , and T are species mass fraction, specific enthalpy, heat of formation at a reference temperature T_0 , isobaric specific heat, gas temperature of the previous iteration, and the unknown gas temperature, respectively.

Therefore, from Equation 47 one obtains the following expression for the gas temperature T :

$$T = T^* + \left[\frac{h - \sum_i m_i h_i(T^*)}{\sum_i m_i c_{pi}(T^*)} \right] \quad (48)$$

The variation of C_p as a function of temperature is taken as a fourth-order polynomial of temperature as given in Reference 16.

7. Finite-Difference Solution of the Equations.

The numerical solution of the nonlinear, coupled, partial differential equations can be obtained by using finite-difference methods. A numerical solution of the hydrodynamic equations can be obtained by two methods. The earlier approach employed for 2-D flows was the so-called streamline-vorticity method³. Here pressure is replaced from the momentum equations by differentiation. Stream function (ψ) and vorticity (ω) replace the velocity components and the pressure, thus requiring solution of only two instead of three variables: namely, u , v , and p . The equations were solved by a point-by-point successive-substitution procedure. Since ψ and ω are linked at the boundaries by way of the no-slip condition, the ω boundary specification could be done a number of ways leading to considerably different false diffusion levels as recently evaluated by de Vahl, Davis, and Mallinson¹⁷. In addition, problems were encountered in obtaining fully converging solutions with nonuniform grid spacing. Since 1973, AiResearch has used a pressure-velocity (primitive variable) solution approach, which has the following three advantages over the $\psi - \omega$ method:

- It permits computation of variable density flows where ρ depends upon pressure and temperature.

¹⁶Gordon, S., and B. J. McBride, "Computer Program for Calculation of Complex Chemical Equilibrium Compositions, Rocket Performance, Incident and Reflected Shocks and Chapman-Jouquet Detonations", NASA SP-273 (1971).

¹⁷de Vahl, Davis, and G. D. Mallison, "An Evaluation of Upwind and Central Difference Approximations by a Study of Recirculating Flow", Computers and Fluids (1976).

- It allows unsteady flows to be calculated as easily as steady ones.
- It works for 3-D flows as well as 2-D flows, whereas the $\psi - \omega$ method cannot be easily extended.

Many primitive variable solution methods have been put forward by different researchers¹⁸. They vary enormously in complexity, ease of use, efficiency, and applicability. The 3-D combustor performance code is based on the well-tried SIMPLE (Semi-Implicit Method for Pressure-Linked Equations) algorithm of Patankar and Spalding as described in Reference 19. The features of the computer model include:

- Solution of sufficiently general single form differential equations
- Provision for use with different physical models
- Use of pressure and velocities as the main hydrodynamic variables
- Use of the pressure correction technique
- Provision of two coordinate (plane and axisymmetric) systems
- Use of nonuniformly spaced grids

¹⁸Anon., "Proceedings of the Third AIAA Computational Fluid Dynamics Conference", Albuquerque, New Mexico, June 27-28, 1977.

¹⁹Patankar, S. V., "Numerical Prediction of Three-Dimensional Flows", in Studies in Convection: Theory, Measurement and Application, Volume 1, Edited by B. E. Launder, Academic Press, 1975.

- Use of staggered grid with attendant minimum truncation errors
- Derivation of finite-difference equations by integrating the differential equations over finite control volumes and thus ensuring mathematical compatibility between the finite difference and the original differential-equation formulations
- Efficient line-by-line tri-diagonal matrix solution of the difference equations
- Unconditional convergence for all Reynolds numbers
- Provision for under-relaxation

A typical grid node spacing is shown in Figure 5. Finite-difference equations for a node are obtained by integrating the differential equations over a control volume enclosing a grid node. For evaluating the convection and diffusion fluxes through a control volume face, a linear variation (in the direction normal to the face) of the flow properties is assumed. For other purposes, a step-wise variation with discontinuities at control-volume boundaries is assumed. Net rate of flow of ϕ into the control volume around a node P (Figure 5) by convection and diffusion in the x-direction is

$$\begin{aligned}
 & [T_{X-} + (1 - f_{X-}) L_{X-}] \phi_{X-} + [T_{X+} - f_{X+} L_{X+}] \phi_{X+} \\
 & - [T_{X-} - f_{X-} L_{X-} + T_{X+} + (1 - f_{X+}) L_{X+}] \phi_P
 \end{aligned}$$

where

$$\begin{aligned}
 T_X &= \Gamma_{\text{eff}, \phi} A_X / \delta_X \\
 L_X &= \dot{m}_X'' / A_X \\
 A_X &= 0.5 (r_+ + r_-) \Delta Y
 \end{aligned}
 \tag{49}$$

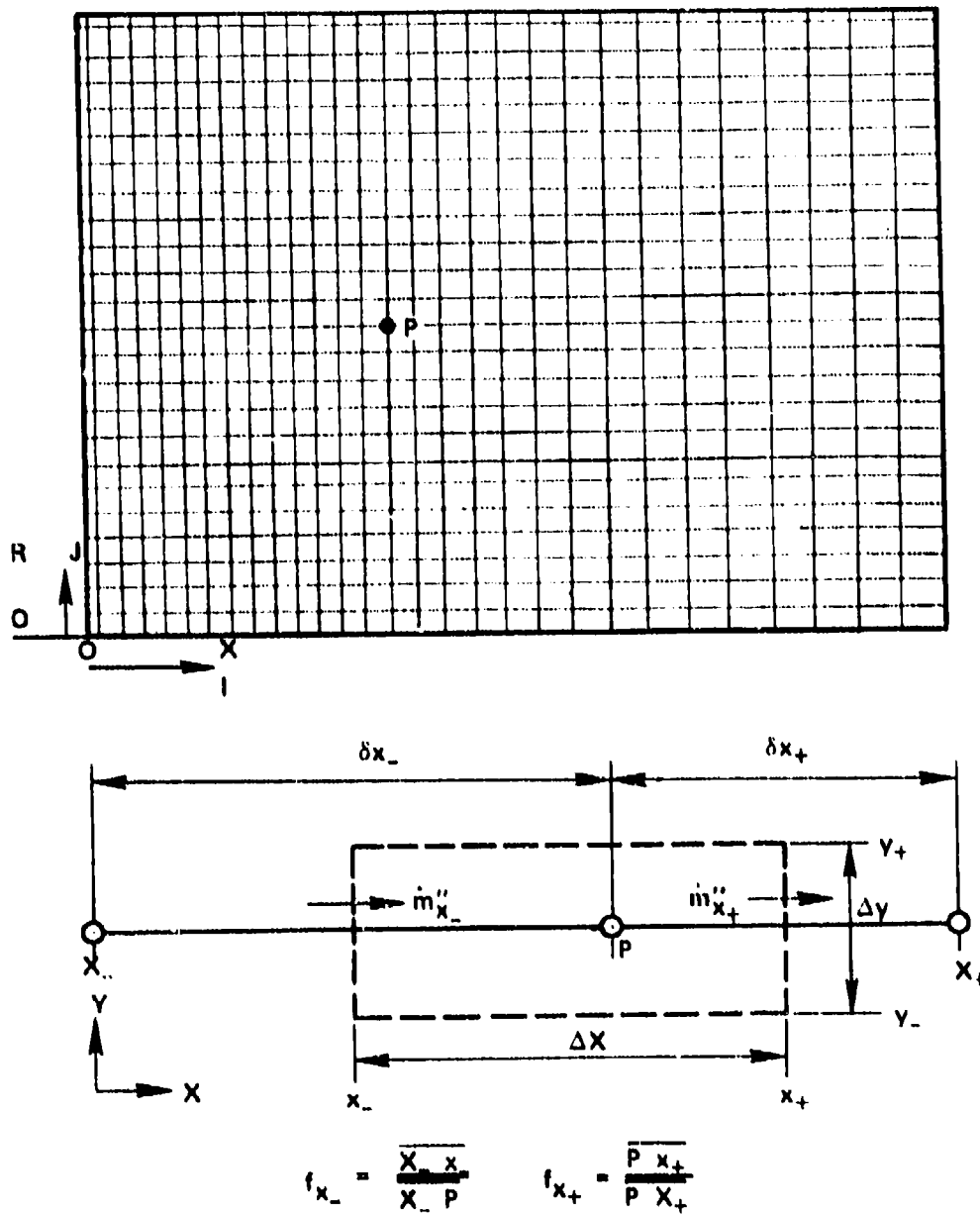


Figure 5. Typical Grid Spacing and Control Value Around a Point P.

Defining $\int \int \int S_v = S_u + S_p \phi_p$, the one-dimensional transport equation for the variable ϕ becomes

$$\begin{aligned} & [T_{X-} + (1 - f_{X-}) L_{X-} + T_{X+} - f_{X+} L_{X+} - S_p] \phi_p \\ & = [T_{X-} + (1 - f_{X-}) L_{X-}] \phi_{X-} + [T_{X+} - f_{X+} L_{X+}] \phi_{X+} + S_u \end{aligned}$$

The above equation was derived based on reasonable assumptions. However, the linear-profile assumption becomes unacceptable when $f_{X+} L_{X+}$ is large compared with T_{X+} because the weighting factor $(T_{X+} - f_{X+} L_{X+})$ then becomes negative, implying an unrealistic physical process through which raising the value of ϕ_{X+} could lower the value of ϕ_p . Therefore, it is assumed that if the convective flow rates (L) are large compared to the diffusion coefficients (T), the diffusion across the control-volume face is zero and the value of ϕ convection is equal to the value at the node on the upwind side of the face. With this assumption, the coefficient $T_{X+} - f_{X+} L_{X+}$ is replaced by $T_{X+}^* - f_{X+} L_{X+}$ where

$$T_{X+}^* = [T_{X+}, - (1 - f_{X+}) L_{X+}, f_{X+} L_{X+}] \quad (50)$$

Here $[a_1, a_2, a_3]$ stands for the largest of the three quantities a_1 , a_2 , and a_3 .

The final finite-difference equation is reduced to

$$\Lambda_p \phi_p = \Lambda_{X+} \phi_{X+} + \Lambda_{X-} \phi_{X-} + \Lambda_{Y+} \phi_{Y+} + \Lambda_{Y-} \phi_{Y-} + \Lambda_{Z+} \phi_{Z+} + \Lambda_{Z-} \phi_{Z-} + S_u \quad (51)$$

The solution of the above equation is obtained by line-by-line relaxation using an efficient tri-diagonal matrix algorithm. By this method, for an x-y plane, a traverse along one direction, say the x-direction, is made with old values for the y-direction nodes. Using this solution as the best estimate, the y-direction is then traversed. The same procedure is repeated for other x-y planes.

8. Boundary Conditions.

The specification of the boundary conditions is done in a number of ways depending upon the problem. For the left inlet boundaries, velocity, density, and turbulence profiles are either experimentally known or estimated. The program can handle any specified profiles. For boundaries of the second kind, where gradients and not the values of the variables are specified, the program uses one of the following two approaches. In the first approach, the boundary value is guessed and continually updated so as to satisfy the given gradient condition. The second approach breaks the link through the boundary to all adjoining external control volumes by first arranging for the finite-difference coefficient connecting the boundary node to an internal node to be zero, and then inserting the correct flux at the boundary as a false source of diffusion and/or convection for that internal node.

At the symmetry plane, the convection and diffusion fluxes are zero. Therefore, the convection coefficient C_{y-} and the exchange coefficient (Γ_{eff}) are made zero at the axis of symmetry. For the exit plane, information about some of the variables is not available. However, since it is the process occurring in the calculation domain that decides values of the variables which the outgoing fluid will carry, there is no need for information at such boundaries. These boundaries are simply treated by making the boundary Γ_{eff} equal to zero. The cyclic boundary conditions are used for the circumferential direction.

The near-wall region is given a special treatment in the program. Since the expression for Γ_{eff} is accurate for turbulent flows only, a means is provided for the inclusion of the correct shear stresses and other fluxes at the wall. Therefore, the nodes next to the wall are assigned the following values as per an empirical wall law:

$$\begin{aligned}
y^+ \leq 11.5 & \quad \Gamma_{\phi, \text{wall}} = \frac{\mu}{\sigma_{\phi}} \\
y^+ > 11.5 & \quad \Gamma_{\phi, \text{wall}} = \frac{\mu}{\sigma_{\phi}} \frac{y^+}{\frac{1}{\kappa} \ln(9y^+) + P_{\phi}} \quad (52) \\
y^+ &= \rho k^{\frac{1}{2}} C_D^{\frac{1}{4}} \frac{\delta}{\mu} \\
P_{\phi} &= 9.0 \left(\frac{\sigma}{\sigma_{\text{eff}}} - 1 \right) \left(\frac{\sigma}{\sigma_{\text{eff}}} \right)^{-\frac{1}{4}}
\end{aligned}$$

Where δ is the normal distance of the wall from the first interior adjacent node. The kinetic energy of turbulence has small diffusion near the wall; hence, Γ_{wall} for k is set equal to zero. Instead of computing Γ_{wall} for ϵ , it is calculated for the near-wall node by assuming a linear variation of the length scale giving the following expression:

$$\epsilon = C_D^{\frac{3}{4}} k^{\frac{3}{2}} / (\kappa \delta)$$

LINER COOLING MODEL

In order to design a durable combustor with conventional materials, the liner-wall temperature levels and gradients must be controlled. Consequently, it is imperative to have a calculation procedure that can be universally used for predicting liner wall temperatures. The wall temperature at a point P in a combustor liner (shown schematically in Figure 6) is determined by energy balance on a control volume around P, i.e.,

$$C_H + R_H = C_C + R_C$$

where C and R denote the heat transfer rate by convection and radiation, respectively. The subscripts H and C correspond to the hot side and cold side of the liner, respectively.

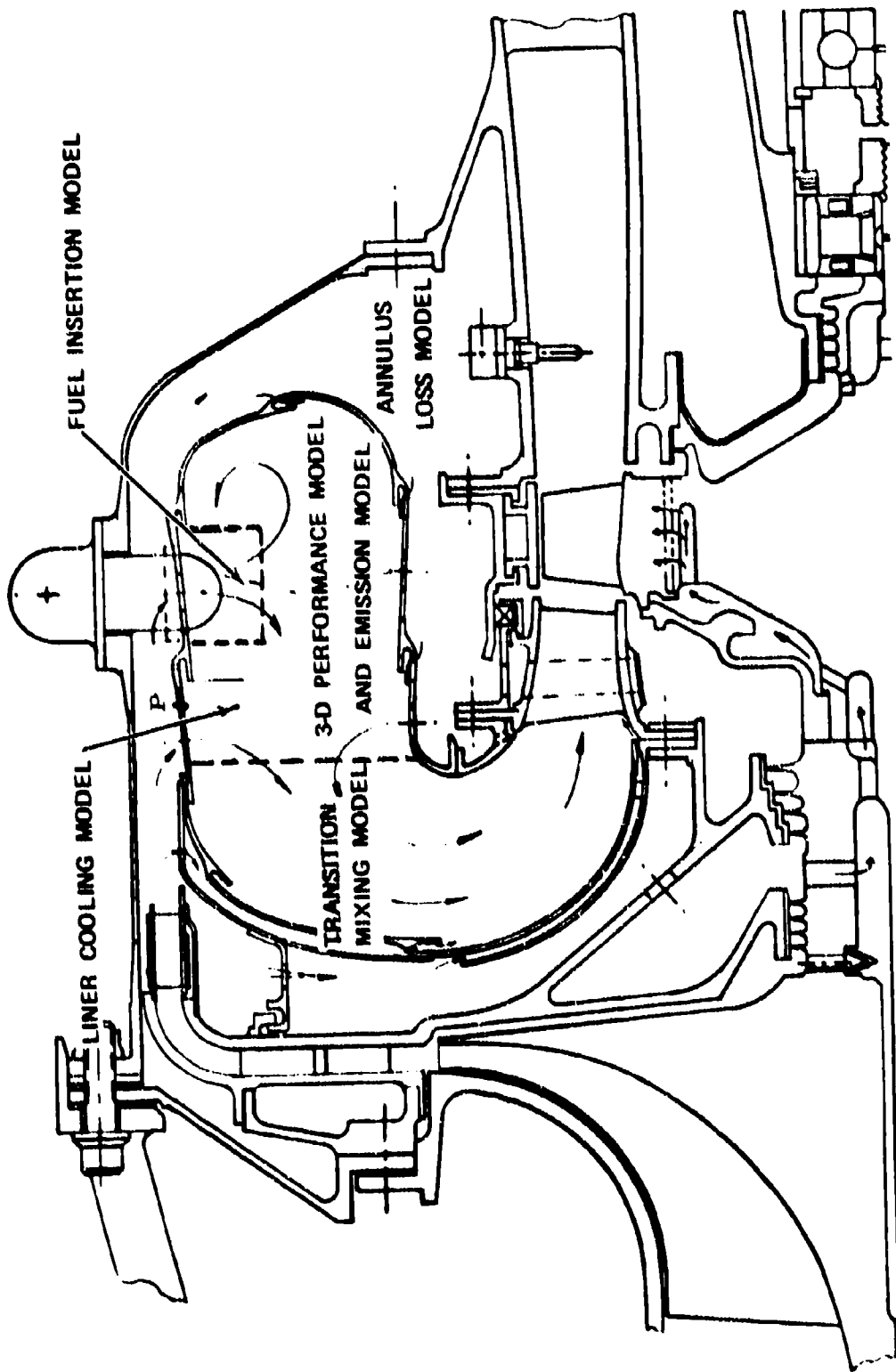


Figure 6. A Schematic of Reverse-Flow Annular Combustor and Application of Analytical Models.

A 2-D parabolic program is used to compute the hot-side convection and radiation heat transfer, the marching direction being x . The following expressions are used for calculating the cold-side heat-transfer rate.

$$C_C = 0.0268 (C_p G)_{an} Re_x^{-0.2} (T_w - T_{an}) \quad (54)$$

$$R_C = \sigma \left[\frac{1}{\frac{1}{\epsilon_w} + \frac{D_L}{D_p} \left(\frac{1}{\epsilon_p} - 1 \right)} \right] (T_w^4 - T_{an}^4) \quad (55)$$

where C_{pan} , G_{an} , and T_{an} are annulus air specific heat, mass velocity, and temperature, respectively. The length Reynolds number is based upon x downstream from the cooling-slot metering orifices. ϵ_w and ϵ_p are the liner-wall and the plenum-wall emissivities; D_L and D_p are the diameters of the liner and plenum, respectively. σ and T_w are the Stefan-Boltzman constant and the liner-wall temperature, respectively.

One major advantage of using algebraic expressions for the cold-side heat-transfer rates is that the appropriate expressions can be used for advanced cooling schemes that increase the heat-transfer rate from the cold side. Consequently, the cooling schemes, such as multiple impingement, extended-surface geometries, and chemically-etched surfaces, can be predicted by using the liner cooling model developed in this program.

Since a 2-D calculation procedure is used for calculating the hot-side heat-transfer rates, the model is strictly applicable to either uncooled liners or the liner walls protected by cooling films. The user will need to make approximations in predicting wall temperatures downstream from discrete radial jets such as the primary and secondary jets.

The 2-D parabolic program solves the governing equations for the following variables:

- Streamwise velocity and swirl velocity
- Turbulence kinetic energy and dissipation model of Jones and Launder²⁰.
- Specific enthalpy
- Unburned fuel, CO, and total fuel appropriate to the two-step kinetic scheme.
- Composite-radiation flux for the two-flux radiation model.
- Five-droplet trajectories

The governing equations, as reduced from the set of equations presented in paragraph B for parabolic flows, are transformed to the following generalized form of transport equations for the von Mises coordinate system²¹.

$$\frac{\partial \phi_j}{\partial x} + (a + b\omega) \frac{d\phi_j}{d\omega} = \frac{\partial}{\partial \omega} \left(c \frac{\partial \phi_j}{\partial \omega} \right) + d_j \quad (56)$$

where

$$a = r_I \dot{m}_I'' / (\Psi_E - \Psi_I) \quad (57)$$

$$b = (r_E \dot{m}_E'' - r_I \dot{m}_I'') / (\Psi_E - \Psi_I) \quad (58)$$

$$c = \frac{r^2 u \mu_{eff}}{(\Psi_E - \Psi_I)^2 \sigma_{j,eff}} \quad (59)$$

$$\omega = (\Psi - \Psi_I) / (\Psi_E - \Psi_I) \quad (60)$$

²⁰Jones, W. P., and B. E. Launder, "The Calculation of Low-Reynolds Number Phenomena with a Two-Equation Model of Turbulence". ASME Paper 72-HT-20, 1971.

²¹Patankar, S. V., and D. B. Spalding, "Heat and Mass Transfer in Boundary Layers", Intertext Books, London; 1970.

Here ϕ_j is a generalized variable, and d_j contains the source/sinks and the other terms in the governing equation that do not fit in the convection and diffusion terms presented in Equation 56. ψ , r , and \dot{m}'' denote streamline, radius, and entrainment rate across the boundaries $\psi = \psi_I$ and $\psi = \psi_E$. The subscripts I and E refer to the inner and outer (external) boundaries of the domain of interest.

The numerical scheme used is a variant of the efficient numerics of Patankar and Spalding as described in Reference 21. A brief description of how the model is used for predicting liner-wall temperatures is given in the following paragraphs.

Consider a typical combustor liner and its predicted isothermal lines for an x-y plane; e.g., in line with primary jets, such as shown in Figure 7. The 3-D combustor-performance model predicted a reverse flow region near the dome, as shown by broken lines. This particular combustor has three cooling slots on the OD liner wall and four cooling slots on the ID wall. The wall-cooling model is used for predicting both inner- and outer-liner wall temperature with initial conditions for the three stations shown, defined by the combustor-performance model. The number of x-y planes to be solved depends upon a particular combustion system. If the predicted combustor internal flow is highly three-dimensional, then one may have to solve as many planes as the number of θ -nodes used in the 3-D computation. However, as the flow field approaches a two-dimensional approximation, one need not analyze more than a few x-y planes to obtain an accurate wall-temperature prediction.

Since the present wall-cooling model is a two-dimensional model, it cannot analyze wall regions near primary, intermediate, and dilution orifices. Consequently, it is advisable to restart the model for each panel with initial conditions as given by the combustor-performance model. Care should be taken in analyzing

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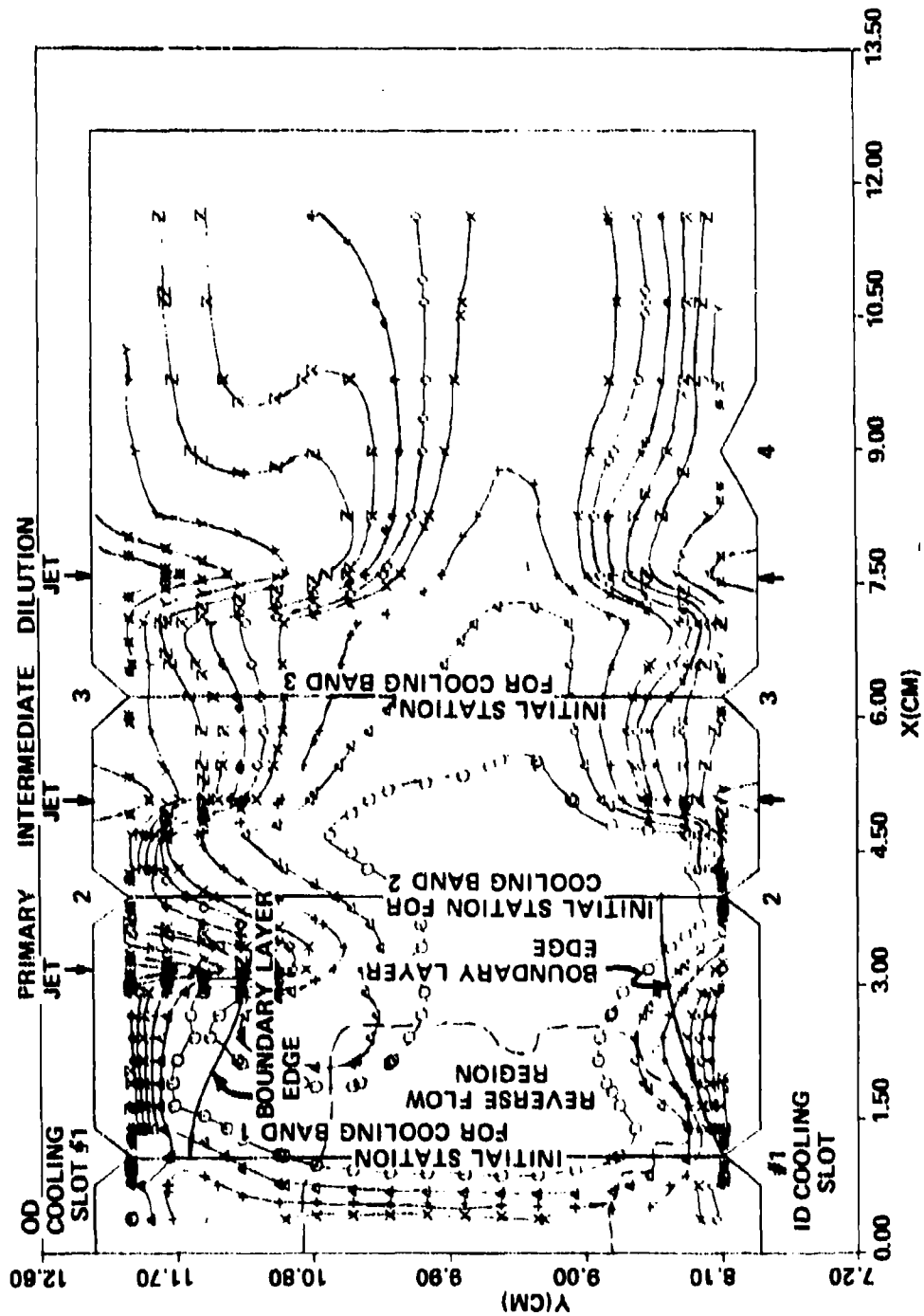


Figure 7. Typical Isothermal Plots of an Annular Combustor Along X-Y Plane In Line With Primary Jet.

the primary panel because of the presence of reverse-flow region. There are two possible approaches for analyzing this section. The first approach, the marching region lies between the liner walls and $u > 0$ with exchange rate specified for $u > 0$ line. In the second approach, one can define a boundary edge for which the edge conditions for dependent variables, including radiation flux, are defined based upon the combustor performance predictions. Then the wall-cooling model is run separately for the inner and outer primary panels.

In order to get more accurate wall temperature predictions, it is imperative to know the precise cooling slot exit conditions. In addition, one must accurately predict the effect of the splash-plate thickness on initial mixing between the cold-stream and main-combustion gases. The 3-D elliptic code can be used with minor modifications to predict the development of the jets exiting from the cooling-slot metering orifices. The effect of liner-pressure drop, orifice size and spacing, slot-lip length, and height on the slot-exit profiles can be analytically predicted. The 3-D elliptic code can also be used to predict the effect of the lip thickness on the initial mixing between hot and cold streams.

TRANSITION-LINER MIXING MODEL

The overall length of a gas turbine engine that employs a centrifugal compressor as the last stage of compression can be minimized by using a reverse-flow combustor. When the engine uses an axial turbine as its high-pressure turbine, a transition liner is needed between the combustor exit and the stator inlet. The transition-liner geometry, as shown in Figure 6, is quite complicated in that combustor exit flow, with mainly axial velocity component flowing from right to left, is bent through a 180-degree turn in order to flow into the turbine stator. The radii of curvature of both the inner- and outer-liner walls vary as a function of distance along the surface.

The flow in a practical transition liner is generally stream wise with little separation. Small regions of separated flows may exist along the ID transition liner surface. But, to insure structural durability of the liner, cooling air is generally injected in these separated areas so that the local liner-temperature levels and gradients are within allowable limits. As burner-exit temperatures increase along with reduction in burner length, cooling-film bands might be used for maintaining transition liner-wall temperature characteristics at an acceptable level.

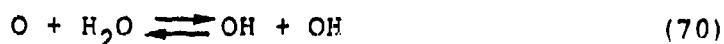
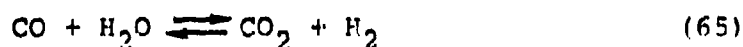
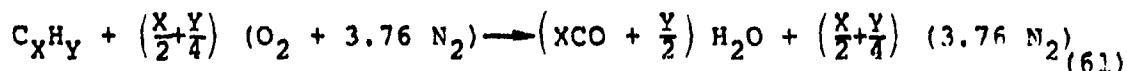
A significant fraction of the mixing of hot streaks with the cooler combustor gases takes place within the transition liner. With the advent of volume-limited turbopropulsion engines, there would be need to predict performance of the transition liner in regard to exhaust-temperature quality. In addition, the transition-liner-wall temperatures must be determined so as to estimate the liner life. A 2-D transition mixing model was therefore developed for this purpose. The program was adopted from the wall-cooling model described in the Liner Cooling Model paragraph. It solves governing equations for streamwise velocity and swirl velocity, specific enthalpy, turbulence kinetic energy, and dissipation. The effect of curvature on radial pressure gradient is taken into account. However, the pressure elliptic effects due to streamline curvature have been neglected.

GASEOUS EMISSIONS MODEL

Both the combustor-performance model and the wall-cooling model use a simple kinetic scheme in that the combustion process is described by two reaction steps, as given by Equations 18 and 19. Such a scheme is manageable for complex computer codes like the combustor-performance model. In addition, the model predicts both unburned fuel and CO which accounts for most of the combustion inefficiency. Since the engineer is generally interested in

estimating combustion efficiency of a given combustor design, the combustor performance model is adequate for the purpose. However, when there is a need for estimating the NO_x emissions, one should preferably use a detailed kinetic scheme so as to predict intermediate species (such as O, N, and OH) that are considered important for the NO production. A number of calculation procedures exist for estimating the NO_x levels; e.g., those described in References 22 and 23. The approach taken here is explained as follows.

The wall-cooling model was modified to incorporate the following 16-step kinetic scheme involving 11 chemical species.



²²Sanborn, J. W., R. S. Reynolds, and H. C. Mongia, "A Quasi-Three-Dimensional Calculation Procedure for Predicting the Performance and Gaseous Emissions of Gas Turbine Combustors", AIAA Paper 76-642, 1976.

²³Mosier, S. A., and R. Roberts, "Low-Power Turbopropulsion Combustor Exhaust Emissions, Volume 3, Analysis", Technical Report AFAPL-TR-73-36, 1974.



A one-step global reaction is assumed for oxidation of fuel to CO as described by Equation 56. This reaction step is similar to the first reaction of the two-step kinetic scheme used in the combustor-performance model and the wall-cooling model. The reaction step is slightly different from that proposed by Edelman²⁴ where his postulation produces H₂ instead of H₂O, assumed here. A set of four reactions are used to describe oxidation of CO. Eight steps are used for reactions involving H₂, O₂ and their dissociation products. Finally, three reaction steps are used for the NO production. Although the program uses the 16-step kinetic scheme, a more extensive kinetic scheme such as that used by Edelman can be incorporated with relative ease.

The fuel-oxidation reaction is controlled by both chemical kinetics and turbulence similar to the scheme used in the two-step kinetic scheme described in paragraph B3. The remaining 15 reaction steps are controlled by chemical kinetics, although the modified eddy-breakup model could be used for these reactions also.

The numerical scheme used in the emission model is slightly different from that used in the wall-cooling model in regard to the way the source term \dot{d}_j of Equation 56 is calculated for the 11 chemical species. For each marching step

²⁴Edelman, R., J. Boccio, and G. Weilerstein, "The Role of Mixing and Kinetics in Combustion Generator NO_x", Paper presented at AIChE Symposium on Control of NO_x Emissions in Direct Combustion Power Sources, 1973.

size ΔX of the parabolic program, d_j^* for the chemical species is computed by using the following 1-D equation

$$\frac{\partial \phi_j}{\partial x} = d_j^* \quad (77)$$

which is obtained by neglecting the cross-stream convection and diffusion terms of Equation 56. Equation 77 is solved for each of the species by taking a number of steps for the distance ΔX . Typically, 50 steps are used for each ΔX . With the source terms for the species now estimated, Equation 56 is then integrated for each of the species over the distance ΔX . Such a modification results in approximately 70-percent reduction in computation time as compared to the numerical scheme used in the wall-cooling model.

FUEL-INSERTION MODEL

It may be recalled that a spray-combustion model is used in the combustor-performance code. This spray-combustion model includes heating, evaporation, and combustion of the spray, as well as the spray trajectories. The code also allows for the exchange of mass, momentum, and energy between the spray and the gas phase. Since a complete solution of the 3-D combustor-performance model takes a long computation time, on the order of three hours on the Cyber 174, an inexpensive calculation procedure was needed for initial selection of the fuel-nozzle characteristics. In addition, such a procedure would allow approximate evaluation of different nozzle designs in a flow field as computed by the combustor-performance model. A fuel-insertion model was therefore developed for this purpose.

Fuel-droplet evaporation rate and heat-transfer rate of the droplet are calculated according to the Priem-Heidmann model as described briefly in the following paragraphs. Vaporization of the droplet \dot{m}_f , lbm/sec is given by

$$\dot{m}_f = A_s K P_{vap}^\alpha \quad (78)$$

$$Nu_m = \frac{2 r_L K}{\rho_{vap} D} = 2 (1 + 0.3 S_c^{1/3} R_e^{1/2}), \quad (79)$$

$$\alpha = \frac{P_\infty}{P_{vap}} \ln \frac{P_\infty}{P_\infty - P_{vap}} \quad (80)$$

where A_s , K , Nu_m , P_{vap} , r_L , ρ_{vap} , D , S_c , R_e , and P_∞ are droplet-surface area, burning-rate constant, Nusselt number for mass transfer, fuel-vapor pressure, droplet radius, fuel-vapor density, diffusivity, Schmidt number, Reynolds number, and surrounding pressure, respectively.

Similarly, heat-transfer rate to the liquid surface α_v , Btu/sec is given by

$$\alpha_v = A_s h (T_\infty - T_L) Z \quad (81)$$

$$Nu_H = \frac{2 h r_L}{k} = 2 (1 + 0.3 P_r^{1/3} R_e^{1/2}) \quad (82)$$

$$Z = \frac{Z}{e^Z - 1} \quad (83)$$

$$Z = \dot{m}_f C_{p,vap} / h A_s$$

where h , Nu_H , k , P_r , and $C_{p,vap}$ are heat-transfer coefficient, Nusselt number for heat transfer, thermal conductivity, Prandtl number, and isobaric-heat capacity of fuel vapor, respectively.

As in the spray combustion model of the combustor performance code, the spray is divided into five discrete droplet sizes. The physical and chemical properties of the jet fuels are varied as a function of the fraction evaporated, as described previously in Paragraph B5.

The following expressions for the spray SMD are currently incorporated in the code. These can be easily changed by the user if desired.

1. Simplex Nozzle.

$$SMD = \frac{225 * W_f^{0.205} \left(\frac{11}{1.5}\right)^{0.3}}{\Delta P_f^{0.354}} \quad (85)$$

2. Simplex Nozzle with Air Assist.

$$SMD = \frac{196 \sqrt{\frac{\sigma \cdot S}{\rho_a}} (11)^{0.095}}{0.438 \left(\frac{W_a}{W_f}\right)^{0.1} v_{aa} \left[0.5 + \frac{v_f}{v_{aa}} - \frac{v_f}{v_{aa}}\right]^{1/2}} \quad (86)$$

3. Duplex Nozzle.

$$SMD = \frac{330 W_f^{0.205} \left(\frac{11}{1.5}\right)^{0.3}}{\left[\frac{\Delta P_p W_{fp} + \Delta P_s W_{fs}}{W_{fp} + W_{fs}}\right]^{0.354}} \quad (87)$$

4. Duplex Nozzle with Air Assist.

$$SMD = \frac{196 \sqrt{\frac{\sigma \cdot S}{\rho_a}} (11)^{0.095}}{0.438 \left(\frac{W_a}{W_f}\right)^{0.1} v_{aa} \left[0.5 + \frac{v_f}{v_{aa}} - \frac{v_f}{v_{aa}}\right]^{1/2}} \quad (88)$$

5. Air-Blast Nozzle.

$$SMD = 1.25 \left(\frac{\sigma \rho_f}{D_f} \right)^{1/2} \frac{(1 + \frac{W_f}{W_{a_n}})}{V_a \cdot \rho_a} + 0.73 \left(\frac{v_f^2}{\rho_a \sigma} \right)^{0.425} \frac{[1 + (\frac{W_f}{W_{a_n}})]^2}{D_f^{0.575}} \quad (89)$$

where:

- W_f = Fuel flow
- μ = Fuel viscosity
- ν = Fuel kinematic viscosity
- ΔP_f = Fuel-pressure drop
- σ = Fuel-surface tension
- S = Fuel-sheet thickness
- ρ_a = Air density
- W_a = Air-assist airflow
- V_{aa} = Air-assist air velocity
- V_f = Fuel velocity
- ΔP_p = Primary fuel-pressure drop
- ΔP_s = Secondary fuel-pressure drop
- W_{fp} = Primary fuel flow
- W_{fs} = Secondary fuel flow
- ρ_f = Fuel density
- D_f = Filming diameter
- W_{a_n} = Air-blast airflow rate

III. DESCRIPTION OF COMPUTER CODES

ANNULUS FLOW MODEL

The annulus-flow model calculates flow conditions around the combustor annulus by solving 1-D fluid-flow equations and provides information regarding annulus axial and tangential velocities, heat transfer from the liner wall, flow rates, jet velocities, jet angles and discharge coefficients of the various liner orifices, as well as the overall liner-pressure drop. Coding logic is provided so that the user may analyze can, and axial-flow and reverse-flow geometries. In addition, options allow the program to calculate the pressure drop for a given inlet flow rate or inlet flow for a given pressure drop. The program will also calculate either the flow through a specified orifice row or the orifice diameters required to pass a specified-flow rate. Finally, a plot, if desired, can be made giving the flow conditions around the combustor annulus and through the liner orifices.

The function of the MAIN program (a computer listing has been provided in Appendix B) is to call subroutine COMANN, which is the main controlling routine, and to perform file manipulations in the case of an axial-flow geometry. For this geometry, one item (usually not known) is the flow split between inner and outer panels. The user inputs essentially two separate cases, one for the OD panel and one for the ID. The program will then iterate on the flow split until the inner- and outer-panel pressure drops are equal. While calculations are being performed on one panel, information about the other is stored on scratch files.

Subroutine COMANN performs the iteration logic and calls the other subroutines as required. Iteration on pressure drop or

flow rate is performed until the calculated flow through the liner orifices agrees to within 0.05 percent of the inlet flow. However, if a solution is not obtained in 20 iterations, the program will stop, as errors in the input or high annulus Mach numbers could make convergence difficult. Of particular importance is the variable RELAX, defined at card C0.31. This is a relaxation parameter used in the convergence logic and has considerable influence on the convergence rate. The simple function provided has worked moderately well for various combustor geometries; however, for complex designs or high annulus mach numbers it is anticipated that the value of RELAX will need to be reduced to obtain a solution.

The names of the remaining subroutines are descriptive of their functions. All data cards are read in subroutine INPUT and then are printed out in subroutine PINPUT. INLET calculates the inlet conditions to the combustor annulus while LENGTH and FLOW perform calculations of annulus flow conditions and orifice flow conditions, respectively. FLOW, in turn, calls JET and DCOEF which calculate the orifice jet velocity and discharge coefficient. Some attention to cards DC.34 to DC.57 in DCOEF is warranted. As there was only qualitative agreement between the measured and calculated discharge coefficients, a constant multiplier was applied to the calculated values. Line printer output is produced in PROUT while the plots are generated in PICTUR and BOXES.

3--D-COMBUSTOR-PERFORMANCE MODEL.

The 3-D performance model is a three-dimensional recirculating-flow program that is capable of analyzing a variety of combustor configurations, including can, can-annular, and annular. The deck solves for the three velocity components, U, V, and W, three species concentrations, including UHC and CO,

turbulence qualities for the $K-\epsilon$ viscosity model, and three radiation fluxes. In addition, the use of primitive variables makes modifications to the boundary conditions easy, allowing the user to analyze complex inlet geometries. Also provided is a subroutine for calculating the trajectories and evaporation rates of a fuel-nozzle spray.

Program MAIN (a computer listing has been provided in Appendix C) is divided into two basic sections. Up to card MA.167, the routine is concerned with reading the input data and converting it to the program's internal units which are Système International (S.I.). The input sequence is covered in paragraph B of Section IV so only the units will be discussed. Cards MA.7 to MA.11 are used to define seven arrays which convert lengths associated with dimensions and lengths associated with velocity, energy, mass, temperature, pressure, and angles respectively. By proper specification in the data statements, the user may employ those input units that are most convenient. The output units are always S.I. From card MA.168 on, MAIN's function is to call the other various routines in their proper sequence.

Subroutine INITIAL performs some preliminary calculations (AL.10 to AL.155), prints the input data (AL.156 to AL.258), and defines the initial conditions and some of the boundary conditions on the various arrays (AL.259 on). In section AL.48 through AL.78, two arrays, JKIN and IKIN, are defined. They merely contain flags which indicate the locations of mass injection points. Cards AL.261 to AL.272 contain logic for the restart option. If Tape 8 from a previous run is saved and then made available for use during a subsequent run, the program will read the initial and boundary conditions from it.

Subroutine ALLMOD contains several entry points which perform miscellaneous calculations pertaining, usually, to the

boundary nodes where modifications to the standard equation are in order. The cyclic nature of the boundary conditions in the θ or K direction is evident in FMOD as well as limits to the fuel and carbon monoxide mass fractions. VELMOD allows the inlet swirl velocity to be increased gradually over a number of iterations and assures that overall continuity is maintained at the exit plane. DENMOD makes alterations to the density at the boundaries to maintain the correct mass-flow rate. GAMOD specifies the wall viscosity values as calculated by the wall functions. SOMAS is used to initialize an array DIVG which is used later in the program. The largest entry point SOMOD contains logic for modifying the equation coefficients and source terms when cooling slots, walls, and droplet evaporation are present. Each variable has its own section and accounts for transfer with the walls and mass addition from the evaporating fuel. SOMODZ deals only with the Z-direction radiation equation and is in a section alone as the data storage is slightly different for this variable.

Subroutine AUX performs the auxiliary calculations for temperature, density, viscosity, and source terms. Entry DENS uses AU.11 to AU.56 to calculate temperature. Cards AU.52 to AU.56 limit the values calculated in order to account for disassociation and early iteration fluctuations. With known temperature, density is then determined from AU.57 to AU.108. VISCO obtains effective viscosity from turbulent kinetic energy and dissipation and calculates y^+ for use by the wall function routine. SOURCE contains all calculations for source terms with the exception of the aforementioned modifications in SOMOD. Again each variable has its own section, with coding that is quite straightforward and requires no explanation.

Subroutine AUXRAD performs the same function as AUX except that it pertains only to the radiation equations.

SPRAY is used to determine the evaporation rate of the fuel-nozzle spray. A large section, from SP.106 to SP.269, deals with locating the droplet, determining free-stream conditions, and handling the situation where the droplet approaches a boundary. Next, various fuel- and free-stream properties are evaluated (to SP.292). The drag forces and time step are then determined and used to obtain new velocities and location. If the droplet is below the boiling temperature, no evaporation occurs (SP.340 to SP.347); but, when the boiling temperature is reached, evaporation rates are calculated, and the appropriate entries to the evaporation array (EVAP) are made. Information concerning momentum changes due to evaporation are also stored in their respective arrays and later (SP.382 to SP.425) on a scratch file for use when the three momentum equations are solved.

The coefficients for each variable are generated and the solution routine called in subroutine STRIDE. First, equations for U, V, and W are handled (ST.117 to ST.632), then the pressure perturbation (P') is obtained (SP.633 to ST.714) and used to correct the velocities (SP.716 to ST.753) so that mass errors are minimized. Then, the remaining variables are solved with the radiation equations having their own special section (ST.915 to ST.937).

STRAD is a subroutine used in the radiation model which performs the same function as STRIDE performed for the other variables.

SOLVE provides a solution to the equations generated in STRIDE. A full three-dimensional solution would be time consuming and would require enormous computer storage. Therefore, an approximate solution is obtained by "sweeping" through the field several times alternately solving along one direction, while holding the values in the other two fixed. The variable ICTDMA

(UV) at S0.36 is used to specify the number of such sweeps. As the program converges, and the variables assume their final values, the solution becomes more and more accurate.

LINER COOLING MODEL

This program is derived from the 2-D parabolic GENMIX program of S.V. Patankar and D.B. Spalding²¹. Modifications have included the addition of a two-equation viscosity model, two-step reaction scheme, two-flux radiation model, plus subroutines for calculating wall temperatures and liquid-fuel-evaporation rates.

The basic geometry for which this program has been geared is continuous inner and outer walls or continuous inner axis of symmetry and outer wall. Other situations may be analyzed provided the proper internal modifications are made to the code.

The MAIN program (a computer listing has been provided in Appendix D) handles several functions, including input, establishment of initial profiles, logic for boundary conditions, calling the additional routines in sequence, and output. The initial section of MAIN (through MA.369) deals with input and initial conditions. Input begins at MA.44 with the case title followed by control indices and grid parameters. More computer storage has been provided than is required for the six species involved in the two-step reaction-scheme, therefore, the extra arrays are zeroed. Various other variables are initialized prior to reading the name list at MA.187. The initial profiles are read from MA.212 to MA.222 and values are assigned to all the arrays from MA.292 to MA.269. The main marching loop (MA.272) begins with the calculation of pressure, temperature, and density (through MA.472). STRIDE(1) called at MA.497 calculates the

²¹Patankar, S. V. and D. B. Spalding.

physical dimension of y from the transformed cross-stream variable ω . The forward step size is next determined along with checks for specified X-locations. The boundary conditions are established between MA.528 and MA.743, and in this deck can be either an inner wall or axis of symmetry and an outer wall. Sections dealing with others are bypassed. The wall temperatures of the inner and outer walls, if required, are determined by the two call statements MA.747 and MA.748, while STRAD (MA.750) is a subroutine used to calculate the radiation flux. Duct geometry and pressure gradient occupy the next section MA.755 to MA.857 and provides two methods for pressure gradient calculation which are selected by IDPDX. When the value is 01, the program uses a guess-correction method, whereas for a value of 02, the program immediately corrects the velocity and pressure fields if the duct area and flow area differ. Entrainment rates are calculated from MA.907 to MA.934 but are not used in any calculations by this code. DROP, a subroutine called at MA.937, calculates the evaporation rates of the liquid-fuel spray. This is followed by STRIDE(2), which performs some preliminary calculations needed prior to solving the equations. The remainder of MAIN is devoted to printout with the exception of STRIDE(3), called at MA.1195 which actually solves the finite difference equations.

Subroutine AUX has two parts; the first (through AU.37) calculates the effective viscosity from the two equation turbulence model, while the second computes the source terms for each equation.

As mentioned above, DROP calculates evaporation rates. Note that the input data for the fuel nozzle is read at AUS.15. With the location of a particular droplet established (AUS.42 to AUS.68), properties of the fuel and free stream are determined (AUS.76 to AUS.94). Some preliminaries are performed before iterative loop AUS.111 to AUS.151 is entered. Calculations are performed until the guessed value of the distance the droplet

travels agrees with the calculated value. With this distance and the droplet velocity known, the time step and evaporation rate can be determined. The proper entries in the evaporation array EVAP are then made at AUS.173 and AUS.190.

STRAD performs all calculations relative to the two-flux radiation model. Modifications to the source terms due to the presence of a wall are made at GA.130 to GA.136. The central difference coefficients are then computed and solved using the standard tri-diagonal algorithm.

WTEMP uses an energy balance on the wall to determine the wall temperature. Cold-side convection, cold-side radiation, hot-side radiation, and hot-side convection are calculated in turn, and a Newton-Raphson iteration procedure is employed to solve the resulting heat-flux equation. The cold-side velocity and temperature are updated at each marching step, accounting for the heat transferred to the annulus.

STRIDE performs the bulk of the numerical calculations and has been documented in literature.

WF is used to evaluate the Couette-flow-equation solutions and to obtain wall shear stress and other transfer data.

PLOTS is a line printer plot routine.

TRANSITION LINER MIXING MODEL

This computer code is derived from the 2-D parabolic GENMIX program of S.V. Patankar and D.B. Spalding. The primary modifications include the ability to have a varying step size across the grid, since, for a given number of marching steps, the distance traveled along the outer-transition liner is considerably greater than along the inner. Other modifications include the addition of $K-\epsilon$ viscosity model.

A computer code that could handle all geometrical configurations would be greatly increased in size; therefore, this deck has been tailored to the geometry of a reverse-flow annular-combustor transition liner. Other configurations can be analyzed if the proper modifications are made to the computer code.

The MAIN program (a computer listing has been provided in Appendix E) handles several functions, including input, establishment of initial profiles, logic for boundary conditions, calling additional routines in sequence, and output. The initial section of MAIN, through card MA.255, contains input and data initialization coding. Note that the x and r values of the boundaries are read at cards MA.64 and MA.65, but that the value of z , the actual marching direction, is computed at MA.74 and MA.77. ISC VE and IPRNT perform the functions their names suggest, determining which variables are solved for and printed. Various constants are initialized in the next few cards. MA.157 is of some significance since it is here that the name list is read, and finally the various profiles are read in and defined. Starting at MA.256, the main marching loop begins with the calculation of pressure, temperature, and density (MA.293 to MA.315). STRIDE(1), called at MA.345, is a subroutine which extracts the physical cross-stream dimension from the transformed cross-stream variable, ω . The forward step size is calculated in the next section (MA.349 through MA.403), which was necessitated by having a smaller step size at the inner boundary than the outer, plus some checks for specified z -locations. Next, the boundary conditions are assigned (MA.406 to MA. 479), and since they are always walls in the transition liner, only those appropriate sections are entered. The actual duct area is determined in section MA.481 through MA.505, plus the area required by the flow. Should these two not agree, compensation in the pressure gradient for the next marching step will be made. The pressure gradient is calculated between MA.507 and MA.598. Two methods are provided and are selected by IDPDX. When the value is 01, the program uses a guess-correction method (MA.539 through MA.543),

whereas for a value of 02, the program immediately corrects the velocity and pressure fields (MA.521 through MA.536). The pressure gradient across the grid due to radius-of-curvature effects is calculated at MA.579 and incorporated into the axial-pressure gradient at MA.595. Statement MA.645 calls AUXO(0), which calculates the effective viscosity from the K- ϵ model. Since there is no entrainment for the geometry employed, cards MA.646 through MA.670 are bypassed. STRIDE(2), called at MA.673, performs some preliminaries necessary prior to the equation solution. The rest of MAIN is devoted to outputs of various types with the exception being MA.902 where STRIDE(3) is called, solving the equations for that marching step.

Subroutine AUX performs two functions: it calculates effective viscosity (up to AU.37) and the source terms for the equations (AU.38 on). Subroutine STRIDE performs the bulk of the numerical calculations and has been heavily documented in literature. Subroutine WF is used to evaluate the Couette-layer-equation solutions near a wall, to extract shear stress and other transfer data needed in the solution of the equations, and finally PLOTS is a line printer plot routine.

EMISSIONS MODEL

The emission model is a 2-D parabolic program derived from the GENMIX deck of S.V. Patankar and D. B. Spalding. The principal modifications include the addition of a 16-step reaction scheme and the ability to handle cooling slots and radial injection orifices.

The MAIN program (a computer listing has been provided in Appendix F) is concerned with input, establishment of initial conditions, logic for boundary conditions calling other subroutines in sequence, and output. The coding is similar to that already described for the liner-cooling model; therefore, only

those items unique to the emissions model will be discussed. Two additional input items are (1) extra specie-input profiles are required (MA.213 to MA.217), and (2) data describing the cooling slots and radial-injector orifices is read at MA.223 to MA.275. To accompany the additional input, there is also logic for the special boundaries, conditions associated with the cooling slots and radial injections MA.562 to MA.618 and MA.659 to MA.715, respectively. When the program reaches the edge of a slot lip, a free boundary is assumed until the flow rate of the slot has been entrained. A similar procedure is used for radial-injection orifices where the boundary is assumed to be a porous wall. All the other subroutines perform the same functions as described in the Liner-Cooling Model paragraph; however, an additional subroutine, AUXS, has been added which calculates the specie source terms and writes them on a scratch file. AUXS solves the same equations as STRIDE except that cross-stream convection and diffusion are omitted. The equations are solved many times using a step size considerably smaller than the main program. In this manner, an estimation of the change in the specie value for the larger main program marching step is obtained from which an average source term over the interval can be calculated. This is then used when the complete equations are solved in STRIDE(3). The first section of AUXS (up to AUS.68) performs data initialization followed by the calculation of forward and backward rate constants (AUS.70 to AUS.95). The main loop is entered next where first the source terms and then the derivatives are determined. Note that the rate expression for the global fuel reaction contains the effect of turbulence (AUS.125) and that the kinetic source term contains a number of variables each raised to a respective power (AUS.123 and AUS.124). These powers, EPU, ERO, etc., are read in through the name list in the MAIN program. A step size, such that the species values do not change excessively, is then selected using the variables TERM1 and TERM2, also part of the name list. Examination of AUS.223 shows that the maximum change allowed during the marching step is the larger

of TERM2 and TERM1 times the upstream specie value. The Equations are then solved and the process is repeated until a distance equal to the main program step size, DX, has been traversed. ISMAX, which is also part of the name list in the MAIN program, is a limit on the maximum number of these steps. The average source term over the interval is then calculated and stored on a scratch file for later use.

FUEL INSERTION MODEL

The function of the fuel-insertion model is to determine the evaporation rates and trajectories of a fuel spray in a two-dimensional flow field. Information concerning the fuel nozzle and flow field are read in, and from these the program calculates the fuel SMD and the trajectories. If desired, a plot of the droplet paths can be made and the evaporation rates saved for use in other programs.

INJECT1 (a computer listing has been provided in Appendix G) is the main program and controls input, output, and the other subroutines. Up to INJ.67 several data statements initialize some fuel and air properties needed in the evaporation calculations. Input data is read to INJ.119 followed by some preliminary calculations. Additional input is read at INJ.187 to INJ.195 if a nonuniform flow field is specified. Calculations of SMD for the particular fuel nozzle type selected begin at INJ.206 continuing to INJ.310. The next section entered (INJ.311 through INJ.435) loops over the five droplet sizes, calculating, in turn, their trajectories. The remainder of INJECT1 provides output and plotting. Subroutine FEVAPC is used to save the evaporation rates for later use, if desired, while AIRPRP interpolates the 2-D nonuniform flow field to obtain the free-stream conditions. Subroutine EVAP performs the majority of the calculations, including the force balances on the droplets so that their trajectories and the evaporation rates can be calculated.

Each of the remaining routines provides some property of the fuel or air required by the calculations.

IV. ILLUSTRATIONS

ANNULUS FLOW MODEL

Figure 8 shows the annulus-loss model example geometry. It is a simple reverse-flow combustor with inner- and outer-panel cooling slots, plunged primary orifices and two dome inlets. For the purposes of analysis, the annulus was divided into elements denoted by the dashed lines. These divisions correspond to places at which mass was extracted or where the annulus was of irregular shape. The elements can be length-type for which skin-friction losses and heat transfer, etc., are calculated or flow-type for which mass extraction is calculated. Thus, the length element (2) is the annulus section between the inlet (1) and (2), and flow element (3) is the OD cooling slot.

Inspection of the input sheets, Figures 9 through 11, shows that Card 1 contains the case title and specifies some control parameters. The annulus-inlet conditions are specified on Card 2 along with the total number of elements used and the inlet-element number. Card 3 is for internal liner flow only and is omitted, whereas Card 4 specifies various constants. It now remains to describe the plenum and liner shape and the various orifices. This is done on the second input sheet with explanations of the various items given in Figure 11. Since, for this case, the flow split between the various orifices was known and not the orifice size, the flow-element cards are of the fixed-flow ratio type (FF). Had the orifice size been known instead, the fixed diameter type (FD) should have been used. If the C_D of a particular orifice row is known, it may be specified as has been done for flow elements 8 and 10. Even though these are actually annular slots in the dome, the program will still calculate an orifice diameter, which is, of course, meaningless; however, the effective and geometric areas are also provided in the output from which the correct annular-slot height can easily

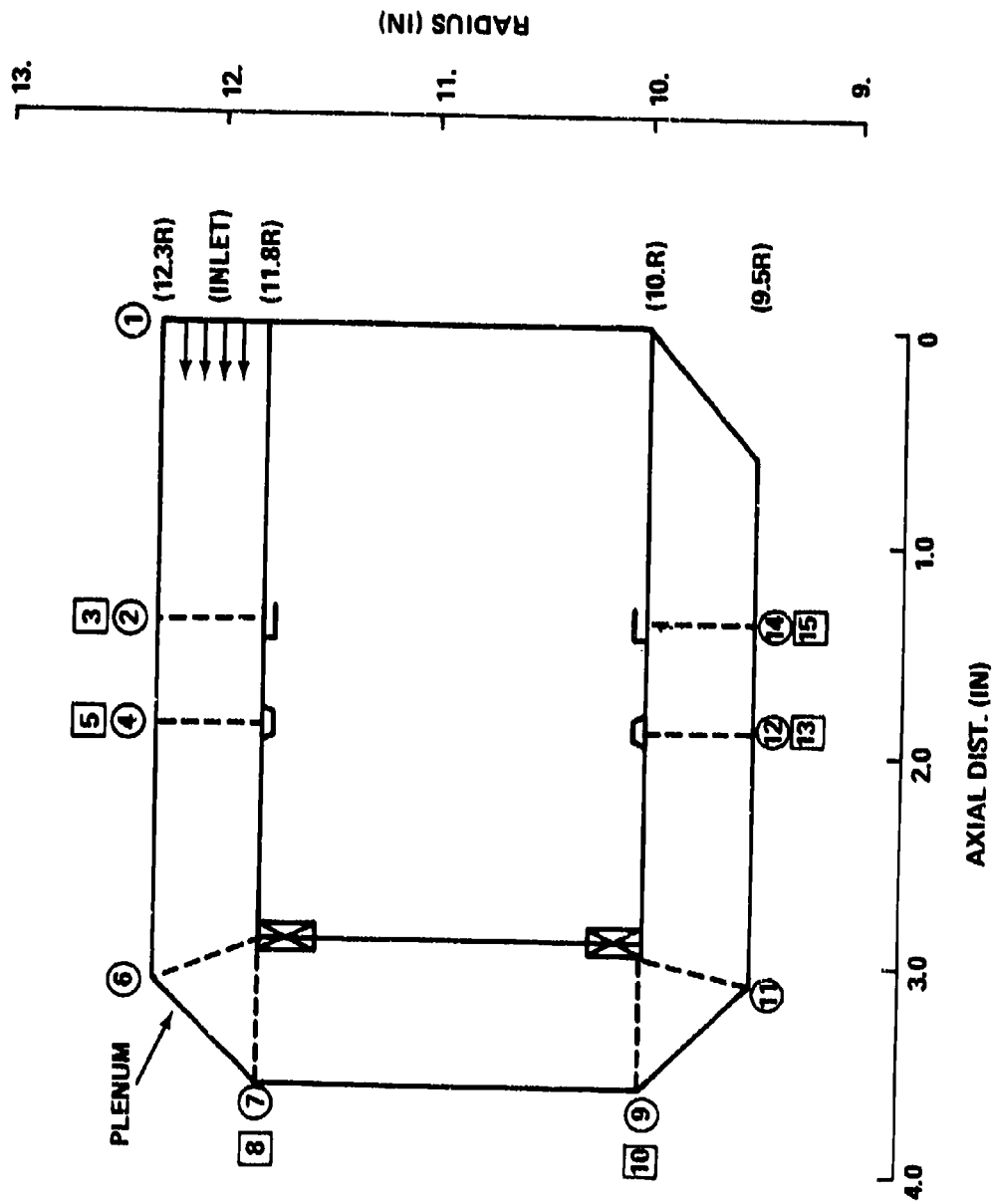


Figure 8. Annulus Loss Model Example Geometry.

1 TITLE - MUST HAVE FOR EACH CASE										ITER	IPIC	IDBUG	
										78	79	80	
1	ANNULUS LOSS MODEL EXAMPLE GEOMETRY										1	1	0
ANNULUS INLET FLOW CARD													
NEL NELI													
1	3	4	5	6	7	8	9	10	11	12	13	14	
	W1	PT1	TT1	BETA1	DP/P								
2	DIS	001	1.6	14.7	104.0	2.0	.03	---	---				
DOME INLET FLOW CARD (REQUIRED FOR INTERNAL LINER FLOW ONLY)													
NELI													
1	2	3	4	5	6	7	8	9	10	11	12	13	
	W	PT	TT	BETA									
3	ID							---	---	---			
CONSTANTS CARD													
1	2	3	4	5	6	7	8	9	10	11	12	13	
	FRIC	BLKF	TANS	CDB	SK	RG	IREAX						
4	C	0.	.83	.1	1.1	1.4	53.3	0.0					

- 1 Title - Run ident appears on printed output and plot
- ITER { = 1 some (or all) holes fixed, inlet flow fixed, iterate to get pressure drop.
= 2 some (or all) holes fixed, pressure drop fixed iterate to get inlet flow (input W1 is first guess).
- IPIC { = 0 no plot
= 1 plot drawn
- IDBUG { = 0 output printed after converged solution
= 1 output printed after each iteration } Use only to de-
= 2 output printed after each element } bugg
non-convergence
- 2 3 NEL = total number of element stations (card 2 only)
NELI = element ID no. (NELEM) at annulus or dome inlet 2 and 3
W1 and W = air flow at inlet stations, lb/sec
PT1 and PT = inlet total pressure, PSIA
TT1 and TT = inlet total temperature, R
Beta1, Beta = swirl angle at inlet
DP, P = total pressure drop, PSIA/PSIA (first guess if ITER = 1) card 2 only
- 4 FRIC { = 0, smooth wall friction factor
= -1, no wall friction
= roughness factor for rough walls
BLKF = annulus effective area factor (= .83 for fully developed turbine flow)
TANS = tangent of flow separation spread angle (.1 recommended)

Figure 9. Annulus Loss Model Input Sheet (Sheet 1 of 2)

CDB = drag coefficient of struts across annulus (1-1.2 RECM)
SK = ratio of air specific heats,
RG = air gas constant
IREAX } = 0. for reverse flow annular or can combustors
 } = 1. for axial flow annular. First data set is for OD
 panel. Program expects a second set for ID panel

CASE TERMINATION

After last card of case:

- o In Column 1, Column 2 blank - case repeated with changes, next card is title card followed by cards with changes from previous run.
- oo In Columns 1 and 2, next card is EOF to quit or new title card followed by all cards for complete new case.

Figure 9. Annulus Loss Model Input Sheet (Sheet 2 of 2)

NELEM									
I		XP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L		XP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L		XP	RP	XL	RL	CL	T LINER	BLK I	DBK I
FF		W/H	N HOLES	NHTYP	CD		I SEP		
FF		W/H	N HOLES	NHTYP	CD	D HOLES	I SEP		
FF		W/H					I SEP		
FF		WTNJ/W	V SET	ANG 1	ABETA		I SEP		

	1	2	345	11	21	31	41	51	61	71	76
1	L	001		0.	12.3	0.	11.8		1960.		
2	L	002		1.4	12.3	1.4	11.8		1960.		
3	FF	003		.15	120.	3.0			1.0		
4	L	004		1.9	12.3	1.9	11.8		1960.		
5	FF	005		.15	30.	2.0			1.0		
6	L	006		3.1	12.3	2.9	11.8		1960.		
7	L	007		3.6	11.8	2.9	11.8		1960.		
8	FF	008		.20	1.0	5.0	.50		1.0		
9	L	009		3.6	1.0	2.9	10.		1960.		
10	FF	010		.20	1.0	5.0	.50		1.0		
11	L	011		3.1	9.5	2.9	10.		1960.		
12	L	012		1.9	9.5	1.9	10.		1960.		
13	FF	013		.15	30.	2.0			1.0		
14	L	014		1.4	9.5	1.4	10.		1960.		
15	FF	015		.15	120.	3.0			1.0		
16		00									
17											
18											
19											
20											
21											
22											
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30											
31											
32											
33											
34											
35											
36											
37											
38											
39											
40											

Figure 10. Sample Work Sheet for Program 117.

NELEM		ALL NUMBERS MUST HAVE DECIMAL POINTS											
1	2	3	4	5	11	21	31	41	51	61	71	76	
L	0	0	1		XP		RP	XL	RL	CL	T LINER	BLK I	DBK I
L	I	0	0	2	XP		RP	XL	RL	CL	T LINER	BLK I	DBK I
L	C	0	0	3	SP		RP	XL	RL	CL	T RISE	--	--
F	F	0	0	4	W/W1		N HOLES	NHTYP	CD	--	I SEP	--	--
F	D	0	0	5	W/W1		N HOLES	NHTYP	CD	D HOLES	I SEP	--	--
F	B	0	1	0	W/W1		--	--	--	--	I SEP	--	--
F	I	0	2	0	W/INJ/W		V JET	ANGJ	ABETA	--	I SEP	--	--

ELEMENT SPECIFICATION

Flow passage is divided into length (L) and flow (F) elements, element numbers, NELEM, can be in arbitrary order, i.e., 10, 1, 3, 4, 16, 30. The cards are stacked in order from inlet to last F because numbers are arbitrary, a new element can be inserted without renumbering other cards.

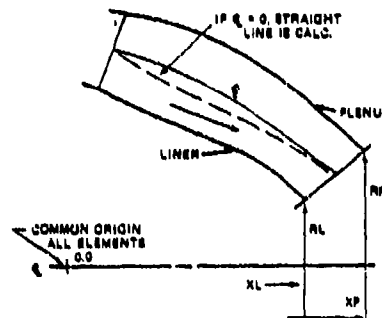
L. LENGTH ELEMENTS All Dimensions in Inches

First L card is annulus inlet (CL = 0) - For both external and internal cases. **LENGTH ELEMENT**
Internal and external flow cases must be run separately

For internal cases, 2nd card is dome inlet (LI) and LC cards are used with T RISE = ΔT due to combustion in this element (no L cards)

For external cases use only L Cards

XL, XP = X COORD to end of element L = Liner
RL, RP = Radius to end of element P = Plenum
(For internal flow XP, RP = OD, XL, RL = ID)
CL = Length of element (optional)
TLIN = Mean wall temp. over CL, °R
if = 0 then TLIN = TTI
BLKI = Frontal Area
Annulus Area
DBKI = Width of strut



F. ORIFICE FLOW ELEMENT

F cards are inserted between L cards at points where flow is extracted. Flow conditions into F elem. are those from upstream L elem.

Types: FF = Fixed Flow Ratio, W/W1
FD = Fixed Orf. Diam. (W/W1 is First Guess)
FB = Bled flow (not included in liner flow)
FI = Internal flow elem. (input to these elements is obtained from an external flow solution)

W/W1 = Orifice Flow/Inlet Flow
NHOLES = No. of Orifices
NHTYP = Hole Type (For CD)
1. Flush Hole, Thin Wall
2. Plunged Hole
3. Cooling Skirt
4. Flush Hole, Thick Wall
5. CD Input
6. Rectangular Hole

CD = DISCH Coefficient (NHTYP = 5 Only)
DHOLES = Hole Diam. (For FD Only)
I SEP = 1, Separation is Reattached
VJET = Jet Injection Velocity, FPS
ANGJ = Jet Injection Angle, Deg.
ABETA = Swirl Angle in Annulus Outside FI Elem.

FLOW ELEMENTS

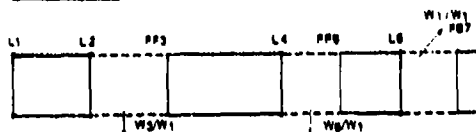


Figure 11. Program 117 Input Data Sheet
Input Format for Element Cards
Sheet 2.

be obtained. Other irregular shaped orifices, such as slotted-dilution holes, for example, can be handled in a similar manner. Card deck termination is specified by the double periods.

The output of the program is shown in Figures 12 through 14. Figure 12 is merely a printout of the input data. Figure 13 gives information of conditions in the annulus at the various stations the user has specified. Figure 14 has information regarding the orifices. Note that the program has calculated orifice-hole diameters and that the flow split is the same as was specified in the input. Overall output parameters such as pressure drop, corrected flow, etc., are also given. The numbers under the heading INPUT FOR SUB BOXES- are associated with the plot subroutine.

COMBUSTOR PERFORMANCE MODEL

The combustor geometry for the 3-D combustor-performance model example is shown in Figure 15. The reverse-flow annular liner has been divided into a grid network consisting of 30 nodes in the axial or X direction, 19 nodes in the radial or y direction, and 13 nodes in the tangential or θ directions. The decision of how large a θ segment to analyze is depending upon where radial planes of symmetry can be found, since the program assumes that there are cyclic boundary conditions in the θ direction. Therefore, any mass leaving the system along the K-13 plane is assumed to reenter the system through the K = 1 plane and vice versa. For this example, uniform grid spacing has been used, although this is obviously not required.

The completed input sheets for the case are shown in Figure 16. Additional input information can be found on the input sheet forms located in Appendix A. Cards 1 and 2 are titles used for printout and case identification. The grid size (30 by 19 by 13) has been entered on Card 3 along with indicators for axisymmetric geometry, K- ϵ viscosity model, kinetic and turbulence controlled

ANNULUS LOSS MODEL: EXAMPLE GEOMETRY

INPUT DATA -
 CALCULATION CONTROLS - SOME (OR ALL) HOLES FIXED SIZE, ITERATE TO FIND PRESSURE DROP (ITER = 1)
 PICTURE WILL BE PLOTTED (PIC = 1)
 OUTPUT PRINTING OCCURS AFTER FINAL SOLUTION ONLY (IDBUS = 0)
 OVERALL PRESSURE DROP - .0450
 ANNULUS EFFECTIVE AREA FACTOR - .8300
 AIR RATIO OF SPECIFIC HEATS - 1.400
 AIR GAS CONSTANT - 53.300
 ANNULUS WALL ROUGHNESS FACTOR - 0.0000
 TANGENT OF SEPARATION SPREAD ANGLE - .1000
 DRAG COEFFICIENT FOR INSERTED BLOCKAGE - 1.1000
 NUMBER OF POSITION ELEMENTS - 15

DIMEN. AT DOWNSTREAM END OF ELEMENT

ELEM NO.	TYPE	LENGTH	OUTER RADIUS	INNER RADIUS	CL ELEMENT LENGTH	MALL R	INSERT AREA FACTOR	INSERT DIAMETER IN.	MOLE DIAM IN.	NUMBER OF HOLES	MOLE TYPE	DISCH COEFF	ORIFICE FLOW / FLOW IN
1	L	0.000	12.300	0.000	11.800	-0.000	1960.0	1.000	1.000				
2	L	1.400	12.300	1.400	11.800	-0.000	1960.0	1.000	1.000	120	3	0.0000	.1500
3	FF	1.900	12.300	1.900	11.800	-0.000	1960.0	1.000	1.000	30	2	0.0000	.1500
4	L	1.900	12.300	1.900	11.800	-0.000	1960.0	1.000	1.000				
5	FF	3.100	12.300	2.900	11.800	-0.000	1960.0	1.000	1.000	1	5	.5000	.2000
6	L	3.600	11.800	2.900	11.800	-0.000	1960.0	1.000	1.000				
7	L	3.600	11.800	2.900	11.800	-0.000	1960.0	1.000	1.000	1	5	.5000	.2000
8	FF	3.600	10.000	2.900	10.000	-0.000	1960.0	1.000	1.000	1	5	.5000	.2000
9	L	3.100	9.500	2.900	10.000	-0.000	1960.0	1.000	1.000				
10	FF	1.900	9.500	1.900	10.000	-0.000	1960.0	1.000	1.000	30	2	0.0000	.1500
11	L	1.900	9.500	1.900	10.000	-0.000	1960.0	1.000	1.000	120	3	0.0000	.1500
12	L	1.400	9.500	1.400	10.000	-0.000	1960.0	1.000	1.000				
13	FF	1.400	9.500	1.400	10.000	-0.000	1960.0	1.000	1.000				
14	L	1.400	9.500	1.400	10.000	-0.000	1960.0	1.000	1.000				
15	FF												

START ITERATION NO. 1
 SOLUTION CONVERGED. TOTAL JET FLOW= 1.6000 FLOW ERROR= -.000000 FINAL PRESSURE DROP= .045000

Figure 12. Annulus Loss Model Output.

ANNULUS LOSS MODEL EXAMPLE GEOMETRY

OUTPUT RESULTS - PAGE 1

ELEM NO	ELEM TYPE	SWIRL ANGLE	FLOW RATE LB/S	MACH NO	FLGM VEL FPS	AXIAL VEL FPS	TANG VEL FPS	TOTAL TEMP, R	STATIC TEMP, R	TOTAL PRESS, PSIA	STATIC PRESS, PSIA	DENSITY LB/FT3	CYMAN HEAD, PSIA	BYN MO /PTOT
1	L	29.00	1.600	.1366	206.09	193.66	70.49	1040.0	1036.5	14.70	14.53	.0379	.174	.0218
2	L	19.76	1.600	.1311	207.75	195.57	70.08	1049.6	1046.0	14.69	14.52	.0375	.175	.0119
3	FF	22.89	1.360	.1136	180.08	165.88	70.08	1049.6	1046.9	14.69	14.56	.0376	.131	.0689
4	L	22.76	1.360	.1137	180.56	166.48	69.90	1053.2	1050.5	14.69	14.56	.0374	.132	.0090
5	FF	27.04	1.120	.0967	153.67	136.86	69.90	1053.2	1051.3	14.69	14.59	.0374	.096	.0045
6	L	28.45	1.120	.0913	145.65	128.05	69.41	1061.7	1059.9	14.66	14.60	.0372	.085	.0058
7	L	30.10	1.120	.0883	140.97	121.94	70.73	1066.2	1062.5	14.69	14.61	.0371	.080	.0056
8	FF	44.52	.800	.0631	100.85	71.87	70.73	1066.2	1063.3	14.69	14.65	.0372	.061	.0028
9	L	43.70	.800	.0740	118.85	85.90	82.14	1076.8	1075.6	14.68	14.65	.0367	.056	.0038
10	FF	57.89	.480	.0403	94.94	51.49	82.14	1076.8	1076.0	14.64	14.65	.0368	.037	.0025
11	L	50.55	.480	.0674	138.49	68.89	83.81	1066.2	1070.2	14.68	14.64	.0366	.047	.0032
12	L	47.66	.480	.0689	111.47	75.04	82.44	1092.1	1091.0	14.68	14.63	.0362	.049	.0033
13	FF	65.51	.240	.0559	90.56	37.49	82.44	1092.1	1091.4	14.68	14.65	.0363	.032	.0022
14	L	64.98	.240	.0551	89.53	37.81	81.16	1101.4	1100.7	14.68	14.65	.0360	.031	.0021
15	FF	89.69	.002	.0499	81.16	.38	81.16	1101.4	1100.8	14.68	14.65	.0360	.026	.0017

Figure 13. Annulus Loss Model Output.

ANNULUS LOSS MODEL EXAMPLE GEOMETRY

OUTPUT RESULTS - PAGE 2

ELEM NO.	ELEM TYPE	CHANNEL WIDTH		ANNULUS GEOMETRIC PARAMETERS		SEPAR AN EFF MAKE		MOLE DISCH		JET ANGLE		DRIF AREA		DRIF AREA		JET VELOCITY		ORIF FLOW		
		IN.	IN.	IN.	IN.	AREA	AREA	IN.	IN.	DEG	DEG	AREA	AREA	EFF	AREA	EFF	AREA	FPS	ORIF	FLOW
1	L	12.050	.500	37.86	29.53	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
2	L	12.050	.500	37.86	29.58	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
3	FF	12.050	.500	37.86	28.94	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
4	L	12.050	.500	37.86	28.97	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
5	FF	12.050	.500	37.86	27.98	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
6	L	12.050	.539	40.77	29.75	1.000	1.000	.343	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
7	L	11.000	.700	51.90	30.81	1.000	1.000	.027	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
8	FF	11.000	.700	51.90	30.70	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
9	L	10.000	.700	43.98	26.38	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
10	FF	10.000	.700	43.98	19.39	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
11	L	9.750	.539	32.99	17.39	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
12	L	9.750	.500	30.63	17.11	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
13	FF	9.750	.500	30.63	10.52	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
14	L	9.750	.500	30.63	10.74	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
15	FF	9.750	.500	30.63	.12	1.000	1.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000

OVERALL FLOW COEFFICIENT = .6282
 INLET CORRECTED FLOW, LB/S = 2.266
 DISCHARGE PRESSURE, PSIA = 14.04
 PRESSURE DROP / FT INLET = .0450
 TOTAL GEOMETRIC AREA, IN2 = 23.981

INPUT FOR SUB BOXES -

2.750	-1.050	-7.050	.500	-7.050	-1.500	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000	0.000
-3.075	2.350	1	-1.050	2.350	2	-2.050	4.575	3	-1.025	2.350	4									
-1.025	4.575	5	.000	2.350	6	1.025	2.350	7	1.025	4.575	8									
2.050	2.350	9	2.050	4.575	10	1.537	4.450	11	.512	-6.450	12									
.512	-6.225	13	-5.113	-6.450	14	-5.113	-5.225	15												

Figure 14. Annulus Loss Model Output.

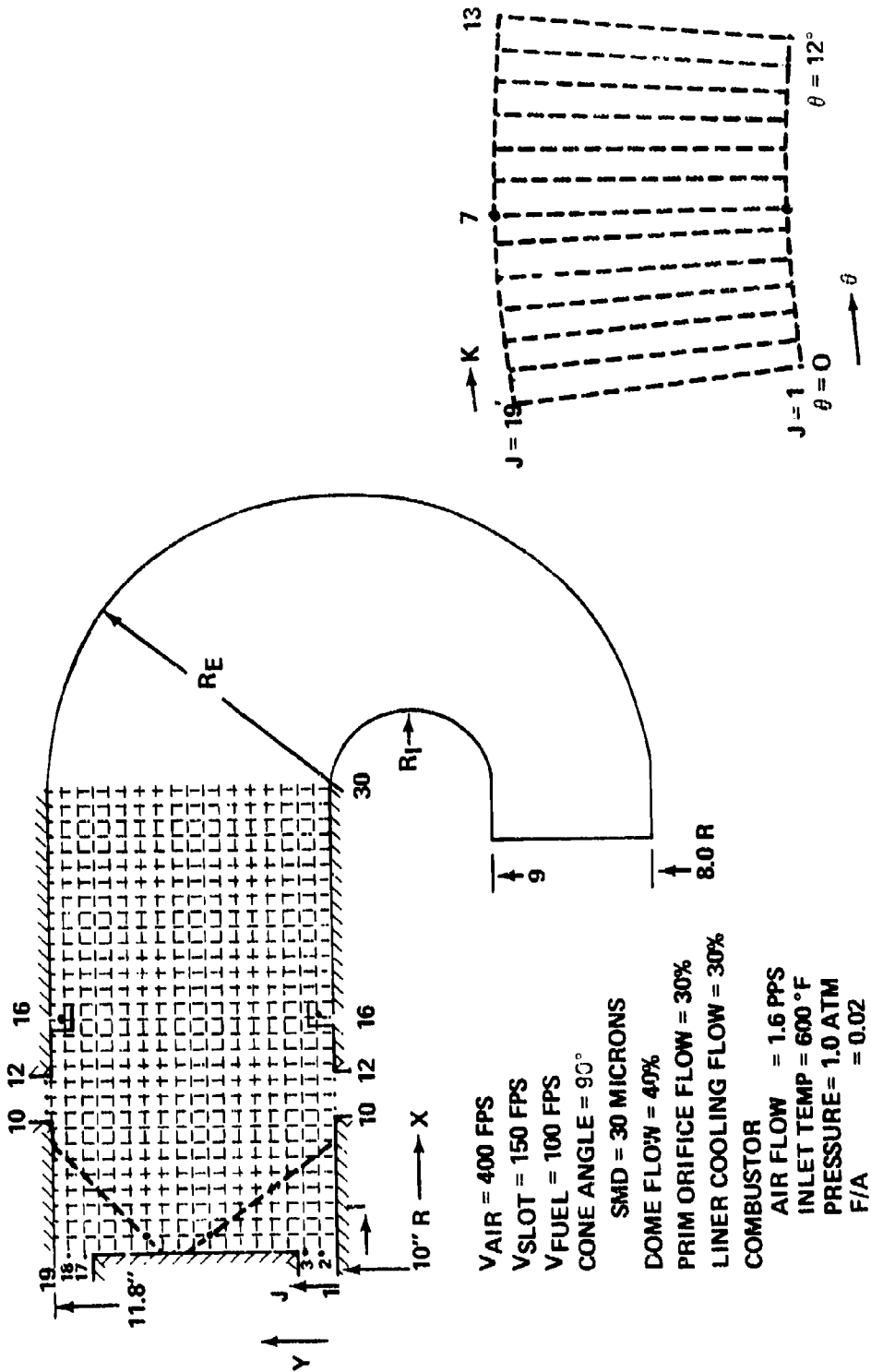


Figure 15. Combustor Geometry for 3-D Combustor-Performance Model.

1	22 VARIABLE TITLE CARDS (4A10) SAME ORDER AS IMPRINT							
2	CASE TITLE CARD (8A10)							
	3D PERFORMANCE MODEL EXAMPLE CASE							
3	LPI	MPI	LPI	IPLAX	MODEL	MODER	IPAR	ITRAD
	30	19	13	02	02	02	02	02
4	IU	MODEN	INTAPE	IDW	IRES			
	02	02	00	01	00			
5	<u>ISOLVE</u> (8(I2, 8X))							
	U	V	W	P'	KE	ϵ	δ	MFU
	01	01	01	01	01	01	01	01
6	MCO	\bar{h}	FX	FY	FZ			
	01	01	01	01	01			
	<u>ICTDMA</u> (8(I2, 8X))							
	U	V	W	P'	KE	ϵ	δ	MFU
	01	01	01	06	01	01	01	01
	MCO	\bar{h}						
	01	01						
7	<u>IMPRINT</u> (8(I2, 8X))							
	U	V	W	PRESS	KE	ϵ_m	δ	MFU
	06	06	06	06	06	06	06	06

Figure 16. 3-D Combustor Performance Model (1 or 6)

TEMP	\bar{n}	Favg	Fx	Fy	Fz	MCO	MH2O
06			06	06	06	06	

M02	MCO2	MN2	μ_{EFF}	DENSITY	EVAP		
			06	06	06		

RELAXATION PARAMETERS (8E10.4)

U	V	W	P'	KE	ϵ	\bar{D}	MFU	
8	.2	.2	.2	1.0	.5	.5	.8	.8

MCO	\bar{n}	Fx	Fy	Fz	PRESS	DENSITY	VISCOS
.8	.8	1.	1.	1.	.5	.2	.1

LAMINAR PRANDTL NUMBERS (8E10.4)

U	V	W	P'	KE	ϵ	\bar{D}	MFU
9	1.	1.	1.	1.	.7	.7	.7

MCO	\bar{n}						
.7	.7						

TURBULENT PRANDTL NUMBERS (8E10.4)

U	V	W	P'	KE	ϵ	\bar{D}	MFU
10	1.	1.	1.		.9	.9	.9

MCO	\bar{n}						
.9	.9						

X-COORDINATES (1-LP1) (8E10.4)

11	0.	.1	.2	.3	.4	.5	.6	.7
----	----	----	----	----	----	----	----	----

.8	.9	1.0	1.1	1.2	1.3	1.4	1.5
----	----	-----	-----	-----	-----	-----	-----

Figure 16. 3-D Combustor Performance Model (2 of 6)

	1.6	1.7	1.8	1.9	2.0	2.1	2.2	2.3
	2.4	2.5	2.6	2.7	2.8	2.9		

RI Y-COORDINATES (2-MP1) (8E10.4)

12

10.	.1	.2	.3	.4	.5	.6	.7
-----	----	----	----	----	----	----	----

.8	.9	1.0	1.1	1.2	1.3	1.4	1.5
----	----	-----	-----	-----	-----	-----	-----

1.6	1.7	1.8					
-----	-----	-----	--	--	--	--	--

Z-COORDINATES (1-NP1) (8E10.4)

13

0.	1.	2.	3.	4.	5.	6.	7.
----	----	----	----	----	----	----	----

8.	9.	10.	11.	12.			
----	----	-----	-----	-----	--	--	--

--	--	--	--	--	--	--	--

IWEI JWIO (2(I2,8X))

14

02	19						
----	----	--	--	--	--	--	--

IWLI VALUES (8(I2,8X)) (SKIP 15 AND 16 IF IWEI = 2)

15

--	--	--	--	--	--	--	--

--	--	--	--	--	--	--	--

Figure 16. 3-D Combustor Performance Model (3 of 6)

16	JWLO VALUES (8 (I2, 8X))							
17	PRESS	DEN	ABSOR	SCATR	AKFAC	ALFAC	(8E10.4)	
	1.0	-	.1	.01	.003	.02		
18	CX	HY	HFU	FUMCO	(8E10.4)			
	1.0	19.28	-49317.	.00001				
19	PREXP1	ARCON1	CR1	PREXP2	ARCON2	CR2	(8E10.4)	
	3.3E+14	27000.	3.0	6.0E+8	12500.	4.0		
	(8E10.4)							
20	C1	C2	CD	AMU	ERROR	TCYLW	TINLW	TLIP
	1.43	1.92	.09	185E-5	.01	1960.	1960.	1560.
	(2 (I3, 7X), 6 (I2, 8X))							
21	LASTEP	IJUMP	JSW1	JSW2	NUINJ	NVINJ		
	150	999	02	03	02	06		
22	USW	VSW	SWNO	AFSW	FSW	TSW	(8E10.4)	
	400.	0.	0.	.02133	0.0	1060		
23	NFNZ	ISPRAY	TFUEL					
	01	9	540					
24	XO	YO	ZO	ALFA	BETA	DELTA	THETA 1	THETA 2
	.05	.9	6.0	90.	-90.	0.0	0.	360.
	(SKIP 24 IF NFNZ = 00)							
	RSP	WF	SMD	VFUEL				
	20.	.001067	30.	100				

Figure 16. 3-D Combustor Performance Model (4 of 6)

(SKIP CARDS 25 → 30 IF NUI NJ = 00)

I - LOCATION OF COOLING SLOTS (8(I2, 8X))
25 | 17 | 17 | | | | | | |

J - LOCATION OF COOLING SLOTS (8(I2, 8X))
26 | 02 | 18 | | | | | | |

AXIAL SLOT VELOCITY (8E10.4)
27 | 150. | 150. | | | | | | |

TANG. SLOT VELOCITY (8E10.4)
28 | 0. | 0. | | | | | | |

SLOT FLOW RATE (8E10.4)
29 | .008 | .008 | | | | | | |

SLOT TEMPERATURE (8E10.4)
30 | 1060. | 1060. | | | | | | |

(SKIP CARDS 31 → 38 IF NVINJ = 00)

I - LOCATION OF RADIAL INJECTION (8(I2, 8X))
31 | 10 | 11 | 12 | 10 | 11 | 12 | | |

J - LOCATION OF RADIAL INJECTION (8(I2, 8X))
32 | 01 | 01 | 01 | 19 | 19 | 19 | | |

K - LOCATION OF RADIAL INJECTION (8(I2, 8X))
33 | 07 | 07 | 07 | 07 | 07 | 07 | | |

INJECTION VELOCITY (8E10.4)
34 | 400. | 400. | 400. | -400. | -400. | -400. | | |

Figure 16. 3-D Combustor Performance Model (5 of 6)

INJECTION TURBULENT KINETIC ENERGY (8E10.4)

35	480.	480.	480.	480.	480.	480.		
----	------	------	------	------	------	------	--	--

INJECTION TURBULENT LENGTH SCALE (8E10.4)

36	.0005	.0005	.0005	.0005	.0005	.0005		
----	-------	-------	-------	-------	-------	-------	--	--

INJECTION FLOW RATE (8E10.4)

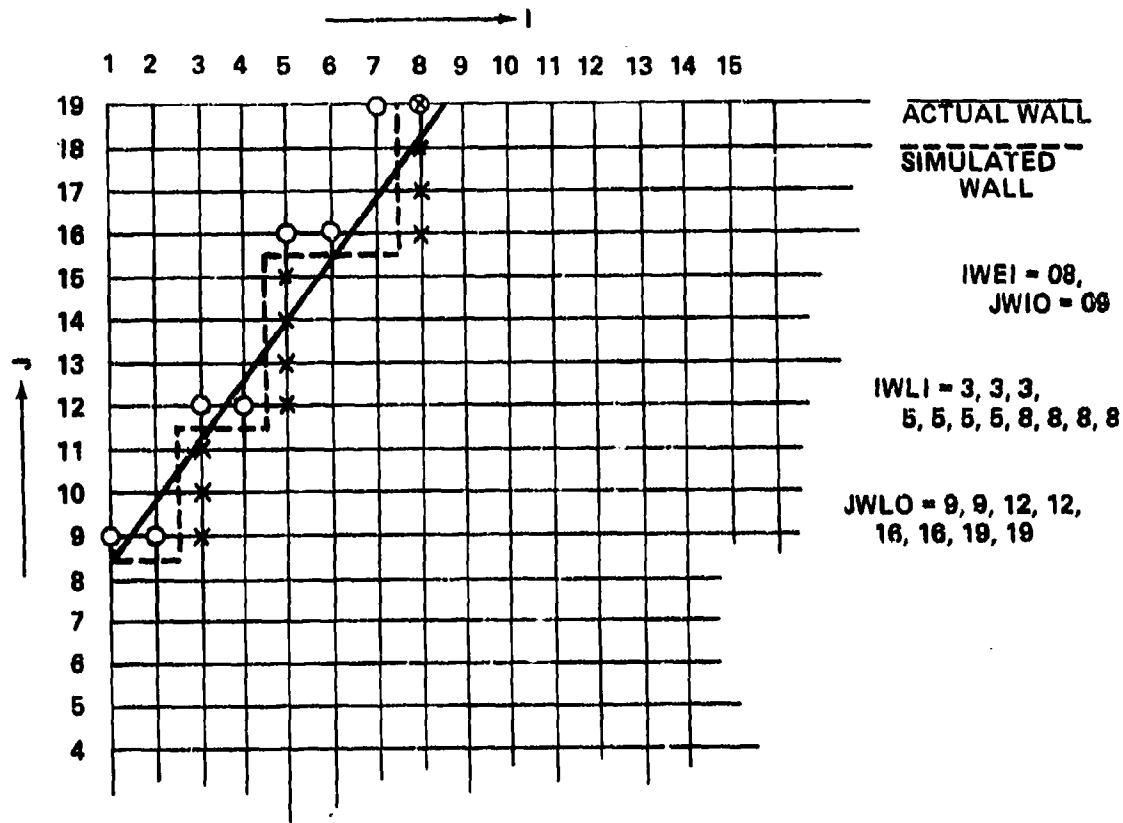
37	.002667	.002667	.002667	.002667	.002667	.002667		
----	---------	---------	---------	---------	---------	---------	--	--

INJECTION TEMPERATURE (8E10.4)

38	1060.	1060.	1060.	1060.	1060.	1060.		
----	-------	-------	-------	-------	-------	-------	--	--

Figure 16. 3-D Combustor Performance Model (6 of 6)

combustion, relative pressure, and radiation to be included, respectively. The use of relative pressure, rather than absolute pressure, merely means that the pressure field is referenced to one particular grid node by subtracting that pressure value from the entire field. On Card 4, the user-selected-input units have been specified. The particular units employed are combinations of convenient ones and are specified by data statements in the main program, at lines MA.9 to MA.11. The user should redefine the arrays for his own input units or use S.I. by specifying IU = 01. Specifying the perfect gas-density calculation, not printing the initial values, the inner boundary as a wall and that this is not a restart, complete the information on Card 4. In order to restart a case, Tape 8 must be saved from the previous run and be made available to the program. Card 5 indicates that all the variables are to be solved. P' is the pressure perturbation used in the solution algorithm, h is stagnation enthalpy, and F_x , F_y , and F_z are the three radiation fluxes. Card 6 shows that all the variables will receive one sweep of the solution routine per iteration except P' , which will have six. The solution of the equations in early iterations is not exact, but this does not present a problem since the coefficients are changing from iteration to iteration anyway. However, it is beneficial to the convergence rate to obtain a more accurate solution to the P' equation. The variables to be printed are specified on Card 7. The 06 merely indicates that every 6th K-plane will be printed. The relaxation parameters specified on Card 8 have been used successfully on a wide range of problems and probably need not be altered. The Prandtl numbers and coordinates, Cards 9 through 13, are fairly self explanatory except that the first field, Card 12, is the radius of the inner boundary and that values of $y(2)$ to $y(MPL)$ are measured from that line. Cards 14 through 16 are used to specify an inclined wall at the inlet. Since there is none for this case, the values of IWEI and JWIO are set equal to 2 and MPI (i.e., 19) respectively, and Cards 15 and 16 are omitted. Were there an inclined wall, Figure 17 gives the necessary information



THE ACTUAL WALL IS SIMULATED BY STAIRSTEPS ALONG THE MIDPOINTS BETWEEN NODES. JWIO IS THE J-NODE WHERE THE INCLINED WALL STARTS, I.E., 09, AND IWEI IS THE I-NODE WHERE THE INCLINED WALL ENDS, I.E., 08. IWLI IS THE 1ST I-NODE INSIDE THE SIMULATED WALL AT A PARTICULAR J-LOCATION. ELEVEN VALUES ARE REQUIRED SINCE THIS IS THE NUMBER OF J-NODES INCLUDED IN THE INCLINED WALL, $J = 9 - 19$, AND ARE MARKED BY *.

JWLO IS THE 1ST J-NODE OUTSIDE THE SIMULATED WALL AT A PARTICULAR I-LOCATION. EIGHT VALUES ARE REQUIRED SINCE THIS IS THE NUMBER OF I-NODES INCLUDED IN THE INCLINED WALL, $I = 1 - 8$, AND ARE MARKED BY ⊙.

Figure 17. Information Necessary to Describe an Inclined Wall.

required to describe it. Cards 17 through 20 need no further explanation other than that given in input sheet forms. A total of 150 iteration steps, along with no intermediate printout, have been specified in the first two fields of Card 21. The basic program contains provisions for only one dome inlet, specified by JSW1 and JSW2; however, the example problem has two. The second inlet was handled by internal modifications to the program in the Subroutine ALLMOD, a process required on nearly every combustor analyzed, as a code which makes provisions for all possible configurations would be extremely complex and lengthy. Note that each primary orifice is simulated by three nodes, making the total of radial-injection points equal to six. Inlet conditions for the dome are provided on Card 22, while Cards 23 and 24 describe the fuel nozzle. A back angle (β) of zero would have the nozzle spraying in a purely tangential and in an increasing θ direction. Positive β would have the spray cone rotated toward the dome; therefore, a value of -90 degrees has the spray cone directed axially along the combustor as is required. Zero down angle (δ) was needed for this example. A positive value would, however, rotate the spray cone toward the inner wall. For geometries that have the entire spray cone in the calculation domain, such as this annular one, THETA1 and THETA2 never change from 0 and 360 degrees or their equivalent in whatever angular input units are selected. Cards 25 through 30 are used to describe the cooling slots, of which two were specified (NUINJ = 2 on Card 21). Note that even though the slots surround I-node 16, the I location is specified as one greater (or 17) due to internal conveniences in the program. Similarly, Cards 31 through 38 specify the radial injection points. Due to the sign convention on V-velocity (positive is in the positive y-direction), injection points on the outer wall have a negative V-velocity component, as shown on the last three fields of Card 34.

The output of the 3-D model is illustrated in Figure 18 (7 sheets). Sheet 1 is a printout of some input data, including fuel and air flow rates, injection velocities, and other important quantities. Note that the output units here are S.I. Sheets 2 and 3 show the u -velocities at $\theta = 6$ deg. (in line with the primary holes) after 150 iterations. The total error in mass, i.e., the sum of the mass error in all the grid cells, was 1.5 percent of the total flow, which was deemed accurate enough for this example. The two dome inlets $J = 2-3$ and $J = 17-18$ can be seen at $I = 2$. The u -velocities are calculated for slightly different control volumes than the other variables. The U -velocity printed at $I = 8$, for example is actually the U -velocity that occurs at the cell boundary between $I = 7$ and $I = 8$. This displacement results in the inlet U -velocity being stored at $I = 2$ rather than $I = 1$. A small recirculation zone exists behind the dome and is terminated by the primary orifices at $I = 10, 11,$ and 12 . Small recirculation regions are also evident behind each jet at $I = 14$, while the presence of the two cooling slots can be seen at $I = 16$ and 17 . Sheets 3 and 4 of Figure 18 show the fuel mass fractions for the same θ plane. The figure also shows the extremely rich regions just behind the dome at $I = 2-4$ near the fuel nozzle. By the time the exit plane is reached, some unburned fuel still remains. Temperature is shown in Sheets 5 and 6. The fuel-rich region behind the dome (seen in Figure 15) exists at a relatively low temperature as the amount of oxygen is very limited. Farther down the combustor where the primary jets have recirculated ($I = 6,7$) one sees temperatures closer to stoichiometric. The primary jets penetrate to the combustor centerline as evidenced by the temperature profile at $I = 12$. These jets produce a colder core with hot regions on either side that extend well past the cooling slots at $I = 16$. These slots provide a cool film that extends to the exit of the combustor. Sheets 6 and 7 show the evaporation rate of liquid fuel. One can clearly see the two sides of the spray cone and that some fuel is impinging on the wall at $I = 9$. Practically all the fuel has evaporated

3D PERFORMANCE MODEL EXAMPLE CASE

I. PHYSICAL INPUT

1. FUEL -

HYDROGEN-CARBON RATIO ----- 1.9280E+00
 MOLECULAR WEIGHT ----- 1.3928E+02 (KG/KMOLE)
 HEAT OF FORMATION ----- -4.9317E+04 (CAL/GMOLE)
 INLET-1 MASS FLOW RATE ----- 0.0 (KG/S)

2. AIR -

PRESSURE ----- 1.0133E+05 (NEM/SQ.M)
 INLET-1 MASS FLOW RATE ----- 9.6753E-03 (KG/S)
 INLET-1 AXIAL VELOCITY ----- 1.2192E+02 (M/S)
 INLET-1 SWIRL NUMBER ----- 0.0

II. GEOMETRICAL INPUT

CHANNEL HEIGHT OF COMBUSTOR ----- 9.1440E-02 (M)
 LENGTH OF COMBUSTOR ----- 7.3660E-02 (M)
 ANGULAR SECTOR ----- 2.0940E-01 (RAD-M)
 INLET-1 FLOW AREA ----- 7.3620E-04 (SQ.M)

III. AIR INJECTIONS

1. FILM COOLING AIR -

SLOT NO	I	J	K	U-VELOCITY (M/S)	V-VELOCITY (M/S)	W-VELOCITY (M/S)	MASS FLOW (KG/S)	FUEL FLOW (KG/S)
1	17	2	000	4.572E+01	2.545E-73	0.0	3.629E-03	
2	17	18	000	4.572E+01	2.545E-73	0.0	3.629E-03	

2. DILUTION AND SECONDARY AIR -

SLOT NO	I	J	K	U-VELOCITY (M/S)	V-VELOCITY (M/S)	W-VELOCITY (M/S)	MASS FLOW (KG/S)	FUEL FLOW (KG/S)
1	10	1	7	2.545E-73	1.219E+02	2.545E-73	1.210E-03	
2	11	1	7	2.545E-73	1.219E+02	2.545E-73	1.210E-03	
3	12	1	7	2.545E-73	1.219E+02	2.545E-73	1.210E-03	
4	10	19	7	2.545E-73	-1.219E+02	2.545E-73	1.210E-03	
5	11	19	7	2.545E-73	-1.219E+02	2.545E-73	1.210E-03	
6	12	19	7	2.545E-73	-1.219E+02	2.545E-73	1.210E-03	

3. FUEL NOZZLES -

X0 (M)	Y0 (M)	Z0 (M-R)	ALFA (RAD)	BETA (RAD)	DELTA (FRAD)	THETA1 (RAD)	THETA2 (RAD)	MSL	MF (KG/S)	SMP (MICRON)	VFUEL (M/S)
1.27E-03	2.29E-02	1.05E-01	1.57E+00	-1.57E+00	0.0	0.0	6.28E+00	2.00E+01	4.84E-04	3.00E+01	3.05E+01

IV. AIR-FUEL BALANCE

TOTAL FUEL FLOW RATE ----- 4.8399E-04 (KG/S)
 TOTAL AIR FLOW RATE ----- 2.4191E-02 (KG/S)
 FUEL TO AIR RATIO ----- 2.0007E-02

Figure 18. 3-D Performance Model Output (Sheet 1 of 7).

V. SOME IMPORTANT QUANTITIES

SPECIFIC HEAT (J/KG-K) 1.157E+03
 ACTIVATION ENERGY (ESEA) 2.7000E+04
 PRE-EXPOSURE ENERGY (E15) 3.300JF*14
 LOSS BREAKUP CONSTANT (L55) 2.0000E+00
 ACTIVATION ENERGY (EMD) 1.2500E+04
 PRE-EXPOSURE ENERGY (EMD) 6.0000E+04
 LOSS BREAKUP CONSTANT (EMD) 4.0000E+00
 TUBE CONSTANT (C1) 1.5000E+00
 TUBE CONSTANT (C2) 1.5000E+00
 TUBE CONSTANT (C3) 1.0000E+02
 ABSORPTION COEFFICIENT 1.0000E+01
 SCATTERING COEFFICIENT 1.0000E+02

U-VELOCITY (M/SEC)

	1	2	3	4	5	6	7	8	9	10	11	12	13
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	1.27E+02	4.47E+01	5.23E+01	5.37E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01
17	1.27E+02	4.47E+01	5.23E+01	5.37E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01
16	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3	1.27E+02	4.47E+01	5.23E+01	5.37E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01
2	1.27E+02	4.47E+01	5.23E+01	5.37E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01	5.28E+01
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

Figure 18. 3-D Performance Model Output (Sheet 2 of 7).

J	I = 26	27	28	29	30
19	0.0	0.0	0.0	0.0	0.0
18	3.32E+01	3.28E+01	3.24E+01	3.25E+01	3.26E+01
17	2.07E+01	2.13E+01	2.20E+01	2.27E+01	2.28E+01
16	2.64E+01	2.63E+01	2.62E+01	2.61E+01	2.63E+01
15	2.94E+01	2.92E+01	2.91E+01	2.91E+01	2.92E+01
14	3.06E+01	3.05E+01	3.05E+01	3.04E+01	3.07E+01
13	3.14E+01	3.14E+01	3.15E+01	3.14E+01	3.17E+01
12	3.25E+01	3.26E+01	3.27E+01	3.27E+01	3.30E+01
11	3.42E+01	3.41E+01	3.42E+01	3.45E+01	3.46E+01
10	3.62E+01	3.75E+01	3.70E+01	3.69E+01	3.70E+01
9	3.63E+01	3.43E+01	3.42E+01	3.44E+01	3.45E+01
8	3.00E+01	3.01E+01	3.02E+01	3.04E+01	3.04E+01
7	2.62E+01	2.63E+01	2.64E+01	2.66E+01	2.87E+01
6	2.71E+01	2.72E+01	2.72E+01	2.73E+01	2.74E+01
5	2.59E+01	2.58E+01	2.58E+01	2.58E+01	2.59E+01
4	2.35E+01	2.35E+01	2.35E+01	2.35E+01	2.36E+01
3	1.94E+01	2.01E+01	2.09E+01	2.15E+01	2.16E+01
2	3.43E+01	3.59E+01	3.55E+01	3.54E+01	3.55E+01
1	0.0	0.0	0.0	0.0	0.0

U-VELOCITY (M/SEC)

FUEL MASS FRACTION

J	I = 1	2	3	4	5	6	7	9	10	11	12
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	0.0	1.37E-02	1.56E-02	1.47E-02	1.44E-02	1.42E-02	1.41E-02	1.55E-02	8.44E-04	4.89E-04	0.0
17	0.0	5.10E-02	4.35E-02	3.58E-02	2.89E-02	2.21E-02	1.61E-02	1.20E-02	1.20E-03	5.25E-04	1.98E-03
16	0.0	2.04E-01	1.31E-01	7.50E-02	5.29E-02	3.88E-02	3.44E-02	1.24E-02	1.65E-03	5.99E-04	1.97E-03
15	0.0	2.28E-01	1.64E-01	1.04E-01	9.98E-02	5.68E-02	5.24E-02	1.87E-02	2.22E-02	7.08E-04	1.81E-03
14	0.0	2.40E-01	2.45E-01	1.76E-01	1.04E-01	8.31E-02	6.04E-02	4.36E-02	3.04E-03	8.47E-04	1.76E-03
13	0.0	2.48E-01	3.03E-01	3.75E-01	2.15E-01	1.08E-01	6.72E-02	2.66E-02	3.66E-03	9.57E-04	1.72E-03
12	0.0	2.62E-01	4.66E-01	4.66E-01	2.63E-01	1.03E-01	6.40E-02	1.36E-02	3.50E-03	9.58E-04	1.69E-03
11	0.0	3.03E-01	2.55E-01	1.53E-01	7.29E-02	2.97E-02	1.17E-02	1.60E-03	2.73E-03	9.58E-04	1.64E-03
9	0.0	3.26E-01	4.95E-01	2.99E-01	1.84E-01	1.13E-01	7.96E-02	4.94E-02	4.80E-03	1.83E-03	9.69E-04
8	0.0	2.92E-01	4.24E-01	4.20E-01	2.38E-01	1.39E-01	7.91E-02	2.36E-02	7.60E-02	1.93E-03	8.91E-04
7	0.0	2.79E-01	3.58E-01	3.86E-01	2.30E-01	1.39E-01	9.92E-02	5.42E-02	8.12E-03	1.93E-03	8.36E-04
6	0.0	2.72E-01	3.13E-01	2.44E-01	1.52E-01	1.18E-01	7.67E-02	4.83E-02	7.31E-03	1.72E-03	7.86E-04
5	0.0	2.66E-01	2.62E-01	1.49E-01	9.98E-02	8.95E-02	7.90E-02	3.88E-02	4.84E-02	1.38E-03	7.41E-04
4	0.0	2.50E-01	1.54E-01	1.23E-01	8.83E-02	6.54E-02	5.94E-02	2.70E-02	2.96E-03	1.12E-03	6.98E-04
3	0.0	7.06E-02	6.26E-02	5.41E-02	4.56E-02	3.78E-02	2.83E-02	1.77E-02	1.73E-03	9.84E-04	6.66E-04
2	0.0	1.82E-01	2.02E-02	2.02E-02	1.99E-02	1.97E-02	1.96E-02	1.46E-02	1.00E-03	9.32E-04	6.51E-04
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

k = 7

Figure 18. 3-D Performance Model Output (Sheet 3 of 7).

J	I = 13	14	15	16	17	18	19	20	21	22	23	24
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	7.80E-03	6.52E-03	1.54E-02	4.72E-03	8.37E-04	9.74E-04	1.01E-03	1.02E-03	1.03E-03	1.02E-03	1.01E-03	1.00E-03
17	6.86E-03	4.18E-03	1.21E-02	9.33E-03	6.28E-03	2.86E-03	2.42E-03	1.59E-03	1.10E-03	7.98E-04	6.07E-04	4.80E-04
16	6.30E-03	2.78E-03	7.75E-03	6.00E-03	4.34E-03	3.10E-03	2.24E-03	1.64E-03	1.22E-03	9.24E-04	7.09E-04	5.51E-04
15	5.84E-03	2.05E-03	4.99E-03	4.07E-03	3.15E-03	2.41E-03	1.84E-03	1.42E-03	1.11E-03	8.70E-04	6.90E-04	5.53E-04
14	5.40E-03	1.70E-03	3.23E-03	2.78E-03	2.27E-03	1.82E-03	1.46E-03	1.17E-03	9.55E-04	7.65E-04	6.23E-04	5.10E-04
13	4.86E-03	2.07E-03	2.10E-03	1.87E-03	1.60E-03	1.35E-03	1.12E-03	9.35E-04	7.77E-04	6.44E-04	5.39E-04	4.51E-04
12	4.08E-03	3.01E-03	1.86E-03	1.38E-03	1.14E-03	9.64E-04	8.24E-04	7.04E-04	6.01E-04	5.13E-04	4.38E-04	3.74E-04
11	2.98E-03	2.13E-03	2.68E-03	2.30E-03	1.94E-03	1.61E-03	1.37E-03	1.15E-03	9.59E-04	8.02E-04	6.78E-04	5.80E-04
10	1.99E-03	1.36E-03	2.06E-03	1.73E-03	1.44E-03	1.21E-03	1.01E-03	8.37E-04	7.07E-04	6.05E-04	5.19E-04	4.50E-04
9	2.89E-03	2.10E-03	1.16E-03	7.55E-04	5.65E-04	4.49E-04	3.66E-04	3.07E-04	2.59E-04	2.19E-04	1.87E-04	1.61E-04
8	3.99E-03	1.52E-03	1.39E-03	1.18E-03	9.88E-04	8.22E-04	6.83E-04	5.88E-04	5.25E-04	4.74E-04	4.27E-04	3.82E-04
7	4.20E-03	1.59E-03	2.22E-03	1.85E-03	1.51E-03	1.22E-03	9.93E-04	8.17E-04	6.91E-04	5.86E-04	4.97E-04	4.19E-04
6	4.62E-03	1.80E-03	3.29E-03	2.69E-03	2.12E-03	1.66E-03	1.32E-03	1.05E-03	8.61E-04	7.25E-04	6.14E-04	5.19E-04
5	4.96E-03	2.48E-03	4.81E-03	3.79E-03	2.87E-03	2.15E-03	1.68E-03	1.26E-03	9.99E-04	8.47E-04	7.25E-04	6.33E-04
4	5.25E-03	3.50E-03	5.37E-03	3.82E-03	2.82E-03	2.08E-03	1.69E-03	1.26E-03	9.99E-04	8.47E-04	7.25E-04	6.33E-04
3	5.51E-03	5.14E-03	1.08E-02	8.11E-03	5.32E-03	3.23E-03	1.99E-03	1.28E-03	8.61E-04	6.11E-04	4.59E-04	3.50E-04
2	5.40E-03	8.63E-03	1.68E-02	5.24E-03	6.47E-04	5.74E-04	6.02E-04	6.15E-04	6.20E-04	6.19E-04	6.14E-04	6.10E-04
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

J	I = 25	26	27	28	29	30
19	0.0	0.0	0.0	0.0	0.0	0.0
18	9.84E-04	9.69E-04	9.51E-04	9.31E-04	8.92E-04	0.0
17	3.93E-04	3.31E-04	2.87E-04	2.55E-04	2.12E-04	0.0
16	4.34E-04	3.66E-04	2.80E-04	2.30E-04	1.78E-04	0.0
15	4.47E-04	3.64E-04	2.98E-04	2.46E-04	1.89E-04	0.0
14	4.19E-04	3.47E-04	2.89E-04	2.42E-04	1.89E-04	0.0
13	3.78E-04	3.18E-04	2.69E-04	2.28E-04	1.82E-04	0.0
12	3.20E-04	2.75E-04	2.36E-04	2.04E-04	1.64E-04	0.0
11	2.99E-04	2.17E-04	1.89E-04	1.66E-04	1.38E-04	0.0
10	1.23E-04	1.04E-04	8.97E-05	7.85E-05	6.92E-05	0.0
9	1.40E-04	1.23E-04	1.09E-04	9.81E-05	8.64E-05	0.0
8	3.41E-04	3.05E-04	2.73E-04	2.47E-04	2.15E-04	0.0
7	6.30E-04	5.51E-04	4.79E-04	4.16E-04	3.39E-04	0.0
6	1.92E-03	8.37E-04	6.80E-04	5.65E-04	4.72E-04	0.0
5	1.37E-03	1.03E-03	7.87E-04	6.05E-04	4.23E-04	0.0
4	3.66E-04	3.04E-04	2.54E-04	2.14E-04	1.67E-04	0.0
3	2.80E-04	2.32E-04	1.98E-04	1.72E-04	1.38E-04	0.0
2	6.02E-04	5.92E-04	5.82E-04	5.70E-04	5.45E-04	0.0
1	0.0	0.0	0.0	0.0	0.0	0.0

FUEL MASS FRACTION

Figure 18. 3-D Performance Model Output (Sheet 4 of 7).

TEMPERATURE °K

K = 7

J	I = 1	2	3	4	5	6	7	8	9	10	11	12
19	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	5.89E+02	5.89E+02	5.89E+02
18	5.89E+02	9.74E+02	9.74E+02	9.74E+02	9.74E+02	9.74E+02	9.74E+02	9.74E+02	9.74E+02	5.89E+02	5.89E+02	5.89E+02
17	5.89E+02	1.49E+03	1.79E+03	1.79E+03	2.04E+03	2.04E+03	2.04E+03	2.04E+03	2.04E+03	6.15E+02	5.90E+02	5.89E+02
16	1.09E+03	1.49E+03	2.09E+03	2.39E+03	2.79E+03	2.79E+03	2.79E+03	2.79E+03	2.79E+03	6.77E+02	5.96E+02	5.89E+02
15	1.09E+03	1.70E+03	1.91E+03	1.91E+03	2.22E+03	2.22E+03	2.22E+03	2.22E+03	2.22E+03	7.54E+02	6.07E+02	5.96E+02
14	1.09E+03	1.69E+03	1.69E+03	1.69E+03	1.75E+03	1.75E+03	1.75E+03	1.75E+03	1.75E+03	8.41E+02	6.21E+02	5.91E+02
13	1.09E+03	1.57E+03	1.35E+03	1.35E+03	1.35E+03	1.35E+03	1.35E+03	1.35E+03	1.35E+03	8.35E+02	6.31E+02	5.93E+02
12	1.09E+03	1.69E+03	1.29E+03	1.19E+03	1.65E+03	2.22E+03	2.09E+03	2.14E+03	1.97E+03	8.75E+02	6.31E+02	5.94E+02
11	1.09E+03	1.37E+03	9.56E+02	1.45E+03	2.05E+03	2.09E+03	2.12E+03	1.74E+03	1.28E+03	8.03E+02	6.31E+02	5.96E+02
10	1.09E+03	1.66E+03	1.49E+03	1.31E+03	1.89E+03	2.16E+03	1.59E+03	1.59E+03	1.20E+03	8.35E+02	6.55E+02	6.04E+02
9	1.09E+03	1.31E+03	1.07E+03	1.36E+03	1.84E+03	2.02E+03	2.02E+03	1.52E+03	1.59E+03	9.57E+02	6.58E+02	6.01E+02
8	1.09E+03	1.66E+03	1.09E+03	1.09E+03	1.54E+03	2.07E+03	2.22E+03	2.02E+03	2.02E+03	9.90E+02	6.58E+02	5.98E+02
7	1.09E+03	1.44E+03	1.24E+03	1.14E+03	1.62E+03	2.07E+03	2.21E+03	2.79E+03	2.08E+03	9.56E+02	6.46E+02	5.95E+02
6	1.09E+03	1.66E+03	1.35E+03	1.53E+03	2.09E+03	2.14E+03	2.22E+03	2.30E+03	2.24E+03	9.57E+02	6.25E+02	5.92E+02
5	1.09E+03	1.66E+03	1.49E+03	1.49E+03	2.00E+03	2.20E+03	2.01E+03	2.01E+03	1.80E+03	6.72E+02	6.08E+02	5.90E+02
4	1.09E+03	1.56E+03	1.75E+03	1.75E+03	2.01E+03	2.12E+03	2.02E+03	1.80E+03	1.80E+03	6.77E+02	5.94E+02	5.89E+02
3	5.89E+02	1.42E+03	1.65E+03	1.55E+03	2.03E+03	2.03E+03	1.95E+03	1.80E+03	1.34E+03	6.15E+02	5.90E+02	5.89E+02
2	5.89E+02	1.65E+03	1.17E+03	9.29E+02	9.32E+02	9.34E+02	9.37E+02	5.49E+02	3.59E+02	5.89E+02	5.89E+02	5.89E+02
1	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	5.89E+02	5.89E+02	5.89E+02

J	I = 13	14	15	16	17	18	19	20	21	22	23	24
19	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03
18	7.52E+02	9.74E+02	9.59E+02	9.59E+02	6.75E+02	7.04E+02	7.21E+02	7.37E+02	7.52E+02	7.67E+02	7.82E+02	7.98E+02
17	7.52E+02	1.07E+03	1.04E+03	1.10E+03	1.17E+03	1.25E+03	1.30E+03	1.33E+03	1.34E+03	1.34E+03	1.34E+03	1.34E+03
16	7.52E+02	1.12E+03	1.13E+03	1.13E+03	1.24E+03	1.29E+03	1.34E+03	1.38E+03	1.42E+03	1.44E+03	1.44E+03	1.47E+03
15	7.37E+02	1.15E+03	1.21E+03	1.25E+03	1.25E+03	1.34E+03	1.38E+03	1.41E+03	1.45E+03	1.48E+03	1.50E+03	1.51E+03
14	7.37E+02	1.14E+03	1.26E+03	1.30E+03	1.34E+03	1.37E+03	1.41E+03	1.44E+03	1.47E+03	1.49E+03	1.51E+03	1.53E+03
13	7.37E+02	1.04E+03	1.24E+03	1.24E+03	1.37E+03	1.40E+03	1.43E+03	1.46E+03	1.48E+03	1.50E+03	1.52E+03	1.53E+03
12	6.78E+02	1.75E+03	1.11E+03	1.24E+03	1.34E+03	1.35E+03	1.42E+03	1.45E+03	1.47E+03	1.49E+03	1.51E+03	1.52E+03
11	6.41E+02	7.25E+02	9.37E+02	9.58E+02	1.04E+03	1.18E+03	1.26E+03	1.32E+03	1.37E+03	1.41E+03	1.44E+03	1.47E+03
10	5.56E+02	7.24E+02	7.96E+02	9.09E+02	9.41E+02	1.01E+03	1.02E+03	1.15E+03	1.21E+03	1.26E+03	1.30E+03	1.34E+03
9	1.12E+03	1.12E+03	1.12E+03	1.27E+03	1.36E+03	1.41E+03	1.45E+03	1.48E+03	1.49E+03	1.50E+03	1.50E+03	1.51E+03
8	7.54E+02	1.14E+03	1.39E+03	1.44E+03	1.49E+03	1.51E+03	1.54E+03	1.56E+03	1.59E+03	1.59E+03	1.60E+03	1.64E+03
7	7.81E+02	1.19E+03	1.35E+03	1.40E+03	1.45E+03	1.52E+03	1.52E+03	1.55E+03	1.58E+03	1.60E+03	1.62E+03	1.64E+03
6	7.74E+02	1.15E+03	1.30E+03	1.34E+03	1.41E+03	1.46E+03	1.51E+03	1.55E+03	1.59E+03	1.62E+03	1.55E+03	1.67E+03
5	7.15E+02	1.13E+03	1.26E+03	1.32E+03	1.38E+03	1.44E+03	1.49E+03	1.54E+03	1.59E+03	1.63E+03	1.66E+03	1.68E+03
4	7.57E+02	1.11E+03	1.24E+03	1.27E+03	1.34E+03	1.41E+03	1.47E+03	1.52E+03	1.55E+03	1.57E+03	1.58E+03	1.62E+03
3	7.50E+02	1.04E+03	1.16E+03	1.21E+03	1.29E+03	1.33E+03	1.43E+03	1.45E+03	1.55E+03	1.57E+03	1.64E+03	1.63E+03
2	7.47E+02	9.74E+02	1.10E+03	5.89E+02	6.61E+02	6.87E+02	7.05E+02	7.21E+02	7.38E+02	7.54E+02	7.71E+02	7.88E+02
1	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03	1.09E+03

Figure 18. 3-D Performance Model Output (Sheet 5 of 7).

TEMPERATURE °K

I =	25	26	27	28	29	30
19	0.0	0.0	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0	0.0	0.0
1	0.0	0.0	0.0	0.0	0.0	0.0

K = 7

I =	1	2	3	4	5	6	7	8	9	10	11	12
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

EVAPORATION RATE (KG/SEC)

Figure 18. 3-D Performance Model Output (Sheet 6 of 7).

J	I = 13	14	15	16	17	18	19	20	21	22	23	24
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	8.33E-07	2.38E-07	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3	1.92E-07	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2	4.56E-07	2.06E-07	1.76E-06	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
J	I = 25	26	27	28	29	30						
19	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

EVAPORATION RATE (KG/SEC)

Figure 18. 3-D Performance Model Output (Sheet 7 of 7).

by $I = 15$ with a few larger droplets persisting at $I = 20$, where they entered from neighboring K-planes.

LINER-COOLING MODEL.

The liner-cooling model was used to predict the wall temperatures downstream of the cooling slots in the combustor shown in Figure 15. Starting with the profiles at the exit of the slot, as predicted by the 3-D performance model, the program marched up to the beginning of the transition liner. At each marching step, the program performed a heat-flux balance at the wall and, thereby, obtained the wall temperature.

The input units, required by the liner-cooling model, are S.I., and since these are the output units of the 3-D performance model, data is easily transferred. Profiles for U-velocity, temperature, turbulent kinetic energy, turbulence length scale, fuel-mass fraction, total fuel, and CO are required along with information about the flow conditions in the inner and outer annuli. Since nearly all the liquid fuel has evaporated at the point where the marching process begins, no input for the droplet evaporation is needed.

The input data sheets (Figure 19, 3 sheets) begin with two cards devoted to case identification (additional input information can be found in the input sheet forms located in Appendix A). Next, 40 cross stream grid cells are specified. It was unlikely that this would be sufficient to obtain a grid-independent solution; however, since this was merely an example case, the maximum dimension allowed in the existing deck was selected. Axisymmetric geometry, solve species equations, two-equation turbulence model, reacting species, wall temperature, and enthalpy calculation completed the specifications on Card 2. Card 3 indicates that the various profiles will be printed every 20 marching steps. On Card 4 one sees that the marching process will start at

CASE TITLE (12A6)

1

LINER COOLING	MODEL	EXAMPLE	CASE			
---------------	-------	---------	------	--	--	--

CASE TITLE (12A6)

KPLANE= 7						
-----------	--	--	--	--	--	--

(8(12, 8X))

N KRAD MASSTR ISWRL MODEL INERT IFLUX ITEMP

2

40	01	01	00	02	02	01	01
----	----	----	----	----	----	----	----

NSTAT NPROF NPLOT ITEST LASTEP (5(15, 5X))

3

99999	00020	99999		99999			
-------	-------	-------	--	-------	--	--	--

XU XULAST FRA XEND XOUT PRESS POWER (8E10.4)

4

.03937	.073	.02	10.	10.	101325	2.0	
--------	------	-----	-----	-----	--------	-----	--

NBP (12)

5

02							
----	--	--	--	--	--	--	--

(8E10.4)

X₁ RI₁ RE₁ X₂ RI₂ RE₂ X₃ RI₃

6

0.	.254	.29972	10.	.254	.29972		
----	------	--------	-----	------	--------	--	--

RE₁ X₄ RI₄ RE₄ X₅ RI₅ RE₅ (8E10.4)

--	--	--	--	--	--	--	--

RA RB RD (8E10.4)

7

0.	.254	.29972					
----	------	--------	--	--	--	--	--

Figure 19. Liner Cooling Model Input Sheet (1 of 3)

NAME LIST

8 INPUT KREAD =1, PREXP1=3.3E+14, ARCONI=27000, S

CRI=3.0, CX=10, MY=19.28

9 UA UD (8E10.4)

0.	0.						
----	----	--	--	--	--	--	--

10 F2A F2D TA TD T_{wall} (8E10.4)

0.	0.	589.	589.	589.			
----	----	------	------	------	--	--	--

11 NIN (I2)

19							
----	--	--	--	--	--	--	--

12 Y VALUES (8E10.4)

--	--	--	--	--	--	--	--

13 U VALUES (8E10.4)

--	--	--	--	--	--	--	--

14 (8E10.4)

--	--	--	--	--	--	--	--

15 TEMPERATURE VALUES (8E10.4)

--	--	--	--	--	--	--	--

16 MFU VALUES (8E10.4)

--	--	--	--	--	--	--	--

17 PHI VALUES (8E10.4)

--	--	--	--	--	--	--	--

Figure 19. Liner Cooling Model Input Sheet (2 of 3)

18	MCO VALUES						(8E10.4)
19	TURBULENT KINETIC ENERGY						(8E10.4)
20	TURBULENCE LENGTH SCALE						(8E10.4)
21	VELA _I	VELA _O	TAN _I	TAN _O	PR _I	PR _O	(8E10.4)
	24.7	58.8	589.	589.	.2413	.3124	
							(I2, 8X, 7E10.4)
22	NFNZ	XDP	YDP	UF	VF	SMD	WF TFUEL
	00						

Figure 19. Liner Cooling Model Input Sheet (3 of 3)

0.03937 meter and proceed to 0.073 meter, using a step size of 0.02 times the grid height. In order to obtain the best estimation of the radiation flux, it is necessary to analyze the entire channel height, which means the inner and outer boundaries are walls. XEND and XOUT are, therefore, set to some large number. The variable POWER is used to distort the grid, since more nodes are required near the wall whose temperature is being predicted. This means, of course, that two runs must be made, one with the grid nodes concentrated near the outer wall (POWER <-1.0), and one with the nodes concentrated near the inner wall (POWER >1.0). If POWER equals 1.0 or -1.0, the resulting grid would be uniform in y.

Two straight walls require only two boundary point specifications, handled by Cards 5 and 6. The three radii specified on Card 7 are also easily understood. The name list is read next with the variables specified explained in the name list input sheet. The values on Cards 9 and 10 pertain to the free boundaries, which are not used on this program; however, entries are made for completeness. Cards 11 through 20 specify the initial profiles. The 19 points used correspond to the 19 radial nodes used by the 3-D performance model and are listed in Table 1. For this example the 3-D output was strictly used; however, it is usually the practice to combine the 3-D profiles with some additional information, if available, describing the profiles inside the slot lip. Card 21 lists data concerning the annulus flow conditions and dimensions while Card 22 is blank due to the absence of liquid fuel.

The output from the program is shown in Figure 20 (3 sheets). Sheet 1 begins with the titles, some control indices, and the values of omega for the transformed cross-stream variable. Information about the annuli is printed next along with the initial profiles. The printing of the variable arrays starts with the value at the inner boundary and continues outward.

TABLE 1. INITIAL PROFILES FOR LINER COOLING EXAMPLE.

Node	Y	U	KE	lm	ϕ	MFU	MCO	TEMP
19	0.04572	45.7	6.27	7.62E-5	0	0	0	589
18	0.04318	45.7	6.27	7.62E-5	0	0	0	589
17	0.04064	26.1	13.5	2.86E-4	0.0273	7.8E-3	0.0190	1135
16	0.03810	28.2	51.9	3.68E-4	0.0270	5.17E-3	0.0205	1210
15	0.03556	25.5	65.7	4.22E-4	0.0270	3.61E-3	0.0198	1270
14	0.03302	22.2	71.7	4.76E-4	0.0269	2.53E-3	0.0182	1320
13	0.03048	20.3	73.3	4.81E-4	0.0265	1.73E-3	0.0158	1355
12	0.02794	21.8	77.5	4.66E-4	0.0239	1.26E-3	0.0141	1300
11	0.02540	40.6	126	3.11E-4	0.0154	1.87E-3	0.00976	1018
10	0.02286	53.4	277	2.74E-4	0.0117	1.86E-3	0.00713	905
9	0.02032	21.0	123	4.72E-4	0.0233	6.6E-4	0.0120	1315
8	0.01778	15.3	83.3	5.17E-4	0.0283	1.08E-3	0.0129	1460
7	0.01524	16.2	76.8	4.87E-4	0.0288	1.68E-3	0.0166	1425
6	0.01270	19.2	71.5	4.52E-4	0.0291	2.41E-3	0.0195	1385
5	0.01016	23.0	65.5	4.15E-4	0.0294	3.33E-3	0.0215	1350
4	0.00762	25.9	52.3	3.62E-4	0.0298	4.60E-3	0.0228	1305
3	0.00508	24.1	12.9	2.87E-4	0.0302	6.72E-3	0.0227	1250
2	0.00254	45.7	6.27	7.62E-5	0	0	0	589
1	0.0000	45.7	6.27	7.62E-5	0	0	0	589

U-Values are for K = 7, I = 17 Other Values are for K = 7
and the avg. of I = 16 and 17.

```

XXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXX
XXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXX
  
```

```

LINER COOLING MODEL EXAMPLE CASE
NAME= 1 I.D. WALL
MODE= 2 I.D.P.R. 1
UD 10.906E+02 0.0
G 2.000E+02 2.999E+02 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 7.333E+02
GROSS 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01
1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01 1.000E+01
  
```

```

OMEGA 3.450E-01 1.692E-01 1.866E-01 1.944E-01 1.337E-02 7.272E-02 3.715E-02 5.354E-02 2.300E-02 2.300E-02
2.981E-01 3.076E-01 3.166E-01 3.256E-01 1.011E-01 2.131E-01 1.044E-01 1.222E-01 2.500E-01 2.500E-01
5.845E-01 6.062E-01 6.293E-01 6.524E-01 4.860E-01 1.450E-01 3.248E-01 4.640E-01 8.820E-01 1.500E+01
  
```

```

BOUNDARY *****
I *****
1 5.200E+01 5.800E+02 2.413E-01
5.200E+01 5.800E+02 3.124E-01
  
```

```

ITER= 3 IAX=1000 IEND=1000 IOUT=1000 IIN=1.0E+1 ICR=1.372E-05 PSI= 0.
RMI= 0.0 PFE= 1.332E-01 PFC= 0.482E+01
RZ= 3.937E-02 UFLUX= 4.660E+00 FLUX(J)= 3.181E+03 2.741E-04 3.335E-03 1.327E-03 0.0 0.0 0.0 0.0 0.0 0.0
  
```

```

MIN= 1 TAUTD= 2.649E-01 ADI(DJ)= 6.906E-03 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0
MAX= 1 TAUMD= 1.543E-02 AJED(J)= -6.163E-04 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0
  
```

```

M1, T5 2.540E-01 3.458E-03 2.858E-04 1.153E-04 2.572E-04 6.472E-04 7.144E-04 1.028E-04 1.400E-03 1.400E-03
2.872E-03 4.822E-03 1.312E-03 4.822E-03 3.052E-03 5.052E-03 5.052E-03 2.548E-03 2.548E-03 2.548E-03
2.950E-02 1.312E-02 3.303E-02 1.906E-02 5.736E-02 3.932E-02 3.932E-02 4.874E-02 4.874E-02 4.874E-02
  
```

```

U 2.300E+01 4.570E+01 5.570E+01 5.570E+01 4.570E+01 4.570E+01 4.570E+01 5.570E+01 4.570E+01 4.570E+01
1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01 1.935E+01
2.102E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01 2.078E+01
  
```

```

FUEL 0.400E-04 2.400E-04 1.000E-03 0.166E-03 0.166E-03 0.166E-03 0.166E-03 0.166E-03 0.166E-03 0.166E-03
2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04 2.400E-04
1.504E-03 1.091E-03 2.435E-03 3.537E-03 4.515E-03 4.515E-03 4.515E-03 4.515E-03 4.515E-03 4.515E-03
  
```

```

CO 0.000E+02 1.821E-02 1.674E-02 0.407E-02 0.407E-02 0.407E-02 0.407E-02 0.407E-02 0.407E-02 0.407E-02
1.498E-02 1.640E-02 1.821E-02 1.821E-02 1.821E-02 1.821E-02 1.821E-02 1.821E-02 1.821E-02 1.821E-02
2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01 2.300E-01
  
```

```

NE 2.270E+00 6.665E+00 1.000E+01 9.270E+00 9.270E+00 9.270E+00 9.270E+00 9.270E+00 9.270E+00 9.270E+00
7.099E+00 7.157E+00 7.200E+00 7.200E+00 7.200E+00 7.200E+00 7.200E+00 7.200E+00 7.200E+00 7.200E+00
7.532E+00 7.580E+00 7.620E+00 7.620E+00 7.620E+00 7.620E+00 7.620E+00 7.620E+00 7.620E+00 7.620E+00
  
```

```

LEM 0.004E-04 1.824E-04 4.676E-04 2.665E-04 2.665E-04 2.665E-04 2.665E-04 2.665E-04 2.665E-04 2.665E-04
5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04 5.580E-04
4.790E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04 4.797E-04
  
```

Figure 20. Liner Cooling Model Output (Sheet 1 of 3).

ITEM	RD	IAH=10000	IBNO=10000	IDUJ=10000	IEFF=1.0000	IFEE=1.0000	IFEF=1.0000	IFEF=1.0000	IFEF=1.0000	IFEF=1.0000	IFEF=1.0000	IFEF=1.0000	IFEF=1.0000
PIVYS	2.549E-01	5.447E-01	1.744E-01	2.945E-04	6.430E-04	4.979E-04	1.437E-03	1.469E-03	1.497E-03	1.527E-03	1.557E-03	1.587E-03	1.617E-03
U	4.295E+01	5.792E+00	1.854E+01	2.718E+01	3.566E+01	4.414E+01	5.262E+01	6.110E+01	6.958E+01	7.806E+01	8.654E+01	9.502E+01	1.035E+02
FUEL	7.873E-03	2.475E-03	6.478E-03	1.045E-02	1.442E-02	1.839E-02	2.236E-02	2.633E-02	3.030E-02	3.427E-02	3.824E-02	4.221E-02	4.618E-02
CD	1.998E-09	1.762E-09	2.495E-09	3.695E-09	5.295E-09	7.495E-09	1.029E-08	1.426E-08	1.944E-08	2.602E-08	3.420E-08	4.418E-08	5.606E-08
ORVGN	2.332E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01	2.320E-01
KE	0.722E+00	1.275E+01	4.065E+01	9.205E+01	1.610E+02	2.410E+02	3.310E+02	4.310E+02	5.410E+02	6.610E+02	7.910E+02	9.310E+02	1.081E+03
LEN	0.884E-04	1.531E-05	7.289E-05	3.712E-05	1.912E-05	9.873E-06	5.075E-06	2.673E-06	1.415E-06	7.593E-07	4.075E-07	2.243E-07	1.221E-07
PHI	2.526E-03	1.077E-02	2.320E-02	4.924E-02	1.035E-01	2.120E-01	4.385E-01	8.985E-01	1.800E+00	3.630E+00	7.260E+00	1.430E+01	2.800E+01
TEMP	2.332E+02	9.279E+02	8.175E+02	7.431E+02	6.894E+02	6.457E+02	6.020E+02	5.583E+02	5.146E+02	4.709E+02	4.272E+02	3.835E+02	3.398E+02
RAD	0.822E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03	1.682E+03
BRUHI	4.156E-05	1.374E-04	1.018E-04	1.325E-04	1.722E-04	2.219E-04	2.816E-04	3.513E-04	4.310E-04	5.207E-04	6.204E-04	7.301E-04	8.498E-04

Figure 20. Liner Cooling Model Output (Sheet 3 of 3).

Since y is always referenced from the inner boundary, $y(1)$ is always zero, so the radius of the inner boundary is printed in its place. It can also be seen that the y values are more closely spaced near the inner boundary, since this was the wall temperature that was calculated during this run. Sheets 2 and 3 of Figure 20 show a printout after 80 marching steps as indicated by ISTEP. The program has traversed to an X-location (XU) of 0.0706 meter, which is very near the end of the combustor. At this position the inner-wall temperature is 958°K. The other profiles are also printed with KE being the turbulent kinetic energy, LEN the turbulence length scale, PHI the total fuel, RAD the radiation-composite flux, and AMU(I) the effective viscosity. It should be noted that the boundary values of the radiation-composite flux are never used and are therefore left at their initial values.

TRANSITION MIXING MODEL

The 3-D performance model was used to predict the flow field up to the end of the combustor liner (Figure 15), but it is not capable of handling the geometric configuration of the transition liner. For that, the transition-mixing model (TMM) is used. Using initial profiles as predicted by the 3-D program, the TMM marches through the transition liner, thereby predicting the temperature distribution at what would be the turbine-stator inlet.

An enlarged drawing of the transition liner is shown in Figure 21. The inlet and exit planes are marked along with several intermediate ones. The location of these is a matter of user choice, but should be enough to simulate the actual liner-wall geometry. The radius of curvature is constant. In this case it is 0.5 inch for the inner wall and 1.9 inches for the outer wall. The input sheets are shown in Figure 22 (3 sheets). (Additional input information can be found in the input sheet forms located in Appendix A.) Card 1 is allocated to case identification. Card 2 shows that 40 cross-stream intervals are selected along with axisymmetric geometry, $K-\epsilon$ viscosity model, and

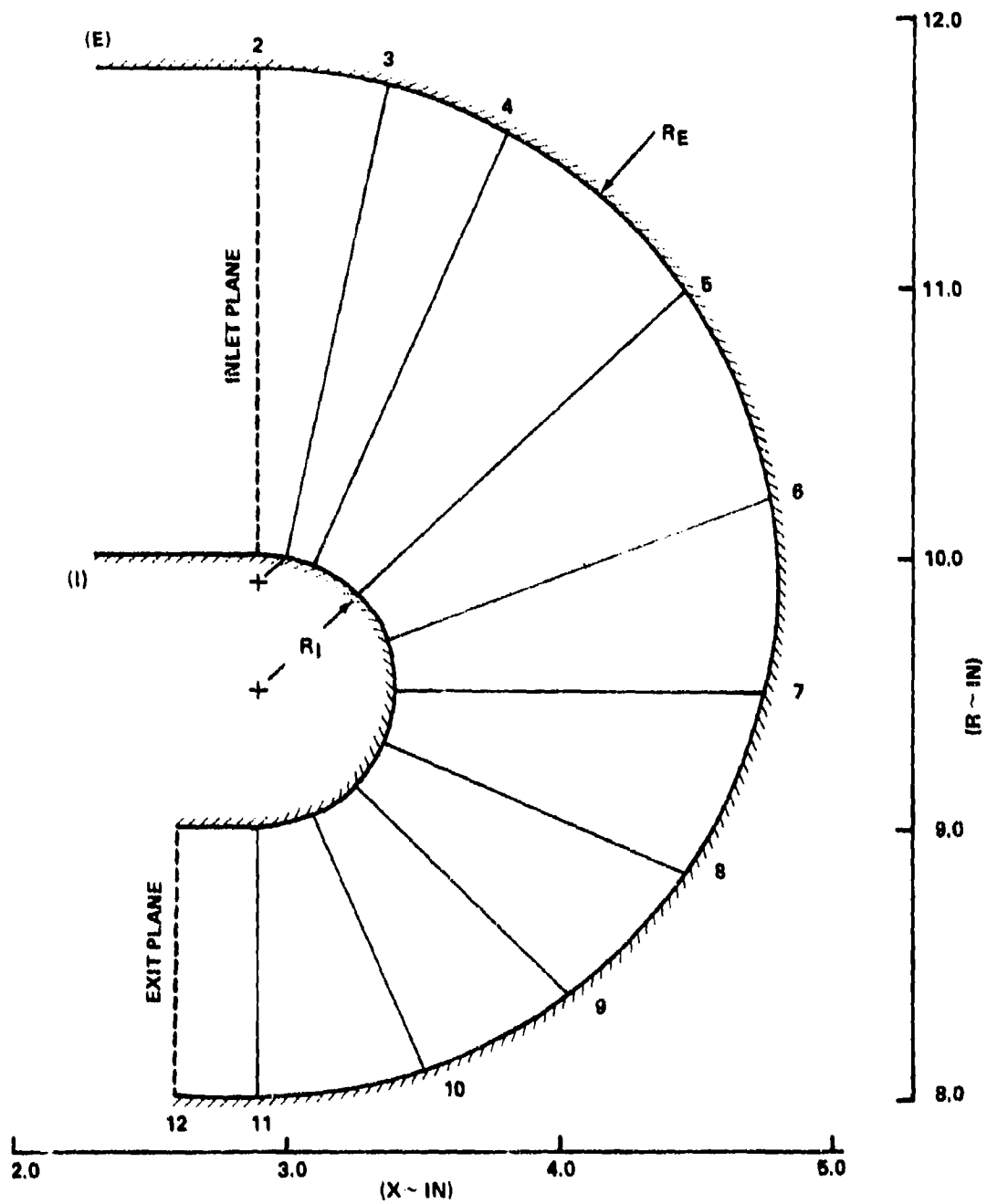


Figure 21. Transition Mixing Example Geometry.

CASE TITLE CARD (12A6)

1 TRANSITION MIXING MODEL EXAMPLE

CASE TITLE CARD (12A6)

KPLANE = 7

(8(12, 8X))

2 N KRAD MASSTR ISWRL MODEL INERT IFLUX ITEMP
40 01 00 00 02 01 00 01

(5(15, 5X))

3 NSTAT NPROF NPLOT ITEST LASTEP
99999 00020 99999 99999

(8E10.4)

4 ZUI ZUE ZULAST FRA ZEND ZOUT PRESS POWER
.07366 .07366 .2316 .02 10. 10. 101325 1.0

(8(12, 8X))

5 NBP ICURV NRCVI NRCVE
12 01 06 06

(8E10.4)

6 RI₁ XI₁ RI₂ XI₂ RI₃ XI₃ RI₄ XI₄...

(8E10.4)

RE₁ XE₁ RE₂ XE₂ RE₃ XE₃ RE₄ XE₄...

(8E10.4)

7 ZI₁ RCVI₁ ZI₂ RCVI₂ ZI₃ RCVI₃ ZI₄ RCVI₄...

Figure 22. Transition Mixing Model Input Sheet (1 of 3)

								(8E10.4)							
ZE ₁		RCVE ₁		ZE ₂		RCVE ₂		ZE ₃		RCVE ₃		ZE ₄		RCVE ₄ ...	
								(8E10.4)							
8	RA	RB	RD												
	0.	.254	.29972												
NAMELIST															
9	SINPUT	KREAD	= 1												S
								(8E10.4)							
10	UA	UD													
	0	0													
								(8E10.4)							
11	F2A	F2D	TA	TD	T _{wall}										
	0	0													
								(I2)							
12	NIN	19													
								(8E10.4)							
13	Y VALUES														
								(8E10.4)							
14	U VALUES														
								(8E10.4)							
16	TEMPERATURE VALUES														

Figure 22. Transition Mixing Model Input Sheet (2 of 3)

17

--	--	--	--	--	--	--	--

18

--	--	--	--	--	--	--	--

19

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20

TURBULENT KINETIC ENERGY						(8E10.4)	

21

TURBULENCE LENGTH SCALE						(8E10.4)	

Figure 22. Transition Mixing Model Input Sheet (3 of 3)

nonisothermal flow. Card 3 will allow a printout of the variable profiles at every 20 marching steps. The initial and final values of Z and the marching direction are placed on Card 4 along with the control on step size. Since the boundaries are always walls, ZOUT and ZEND are set to a large number while POWER equal to 1.0 forces the initial grid to be uniform in y . POWER >1.0 will distribute more nodes near the inner wall while $0 < \text{POWER} < 1.0$ will distort the grid toward the outer wall. Next, the number of boundary points and number of radius-of-curvature points are specified separated by a flag to indicate that radius-of-curvature effects are to be included. Card set 6 and 7 reads the actual boundary and curvature values which are listed in Table 2. The values listed are in inches and had to be converted to meters since the TMM requires S.I. units. The initial and final values of radius-of-curvature were just chosen to be a large (1000m) number. Cards 8 and 9 are quite obvious with the name-list variables listed with the input forms. Cards 10 and 11 deal with "free" boundaries, which are not used for this program, so the values listed are only for completeness. Card 12 indicates that 19 points on the input initial profile were used and correspond to the 19 radial nodes employed by the 3-D program. The various profiles are read on Cards 13 to 21 and are listed in Table 3. These profiles are merely the exit plane profiles for $\theta = 6$ degrees (in line with the primary orifices) as obtained from the 3-D output.

The output of the TMMs is illustrated in Figure 23 (4 sheets). It begins with a list of control indices and important quantities followed by ω , the transformed cross-stream variable. The specified boundary values of X and R for the inner and outer wall along with the value of Z , the distance along the wall, are listed next. Following these are the values of radius-of-curvature and the initial profiles. Printing of the variable arrays starts with the value at the inner boundary and continues outward. Since y is always referenced from the inner boundary,

TABLE 2. TRANSITION MIXING MODEL GEOMETRY INPUT.

<u>Point</u>	<u>XI</u>	<u>RI</u>	<u>XE</u>	<u>RE</u>
1	0.0	10.0	0.0	11.8
2	2.9	10.0	2.9	11.8
3	3.01	9.99	3.37	11.73
4	3.10	9.96	3.80	11.56
5	3.26	9.85	4.45	10.98
6	3.37	9.68	4.76	10.20
7	3.40	9.50	4.75	9.50
8	3.37	9.30	4.45	8.79
9	3.26	9.14	4.02	8.38
10	3.10	9.04	3.51	8.10
11	2.90	9.00	2.90	8.00
12	2.60	9.00	2.60	8.00

<u>Point</u>	<u>ZI</u>	<u>RCVI</u>	<u>ZE</u>	<u>RCVE</u>
1	0.0	39370.0	0.0	39370.0
2	2.9	39370.0	2.9	39370.0
3	3.01	0.5	3.37	1.9
4	4.27	0.5	8.19	1.9
5	4.47	39370.0	8.81	39370.0
6	4.77	39370.0	9.11	39370.0

TABLE 3. INITIAL PROFILES (SI UNITS) FOR TRANSITION MIXING
EXAMPLE

Node	r	u	KE	lm	Temp.
19	0.04572	32.6	15.6	5.86E-4	879
18	0.04318	32.6	15.6	5.86E-4	879
17	0.04064	22.8	18.0	6.03E-4	1310
16	0.03810	26.3	27.0	6.16E-4	1470
15	0.03556	29.2	30.3	6.50E-4	1530
14	0.03302	30.7	30.6	6.64E-4	1560
13	0.03048	31.7	29.6	6.65E-4	1570
12	0.02794	33.0	28.7	6.59E-4	1560
11	0.02540	34.6	29.9	6.42E-4	1530
10	0.02286	37.0	61.8	4.65E-4	1470
9	0.02032	34.5	42.9	6.24E-4	1550
8	0.01778	30.6	27.0	6.58E-4	1660
7	0.01524	28.7	26.4	6.57E-4	1700
6	0.01270	27.4	27.8	6.59E-4	1720
5	0.01016	25.9	28.3	6.49E-4	1690
4	0.00762	23.6	25.3	6.21E-4	1570
3	0.00508	21.6	20.4	6.10E-4	1370
2	0.00254	33.5	16.4	5.88E-4	879
1	0.00000	33.5	16.4	5.88E-4	879

K = 7

DATE	TIME	WAVELENGTH	TEMPERATURE	INDEX OF REFRACTION	COEFFICIENT OF REFRACTION	SCATTERING COEFFICIENT	EXTINCTION COEFFICIENT	TRANSMISSION COEFFICIENT
1964	11:11	4.000	1.111	0.000	0.000	0.000	0.000	1.000
1964	11:11	4.000	1.111	0.000	0.000	0.000	0.000	1.000
1964	11:11	4.000	1.111	0.000	0.000	0.000	0.000	1.000

Figure 23. Transition Mixing Model Output (Sheet 2 of 4).

LEM SC	7.311E-04	5.265E-02	3.205E-04	6.201E-04	5.971E-04	7.132E-04	7.205E-04	7.395E-04	7.511E-04	7.554E-04
	1.622E-03	1.017E-03	1.008E-03	9.579E-04	9.590E-04	9.590E-04	9.590E-04	9.590E-04	9.590E-04	1.022E-03
TEMP	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03	1.572E+03
	1.415E+03	1.355E+03	1.295E+03	1.235E+03	1.175E+03	1.115E+03	1.055E+03	9.95E+02	9.35E+02	8.75E+02
PRESS	1.000E+02	2.557E+02	3.149E+02	3.741E+02	4.333E+02	4.925E+02	5.517E+02	6.109E+02	6.701E+02	7.293E+02
	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02	1.505E+02
AMULL	4.126E-05	5.210E-05	2.462E-04	6.545E-04	4.276E-04	4.135E-04	4.077E-04	4.020E-04	3.963E-04	3.906E-04
	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04	5.679E-04
ISLEP	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00
	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00
PIV'S	2.200E-01	2.474E-03	7.205E-04	1.575E-03	2.118E-03	2.660E-03	3.000E-03	3.444E-03	3.887E-03	4.331E-03
	2.474E-03	7.205E-04	1.575E-03	2.118E-03	2.660E-03	3.000E-03	3.444E-03	3.887E-03	4.331E-03	4.774E-03
ISLEP	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00
	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00	1.000E+00
PIV'S	2.200E-01	2.474E-03	7.205E-04	1.575E-03	2.118E-03	2.660E-03	3.000E-03	3.444E-03	3.887E-03	4.331E-03
	2.474E-03	7.205E-04	1.575E-03	2.118E-03	2.660E-03	3.000E-03	3.444E-03	3.887E-03	4.331E-03	4.774E-03
LEM SC	1.055E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01
	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01	1.555E+01
TEMP	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03
	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03	1.200E+03
PRESS	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02
	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02	1.000E+02
AMULL	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05
	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05	4.000E-05

Figure 23. Transition Mixing Model Output (Sheet 3 of 4).

Y(1) is always zero; therefore, the radius of the inner boundary is printed instead. Note that the pressure variation across the grid (PRESS) is essentially zero due to the large values of radius-of-curvature (RCVI, RCVE) specified at the inlet plane. Sheets 2 and 3 of Figure 23 show the printout after 160 marching steps. Along the outer boundary, the program has marched to 0.1639 meter (ZUE), but the inner boundary is only to 0.09497 meter (ZUI). At this point, there exists a considerable radial pressure gradient with the inner wall at 744 N/m^2 lower pressure than the outer. In addition, the velocity profile is distorted so that the maximum is very near the inner wall. Sheets 3 and 4 of Figure 23 show the output at the exit plane. The radial-pressure gradient disappears as the radius-of-curvature effects are no longer present. A plot of velocity and temperature is also made. The top line of the plot corresponds to the inner boundary while the bottom line is the outer. However, it must be remembered that the inner boundary here is located at what would be the stator-blade tip and the outer at what would be the blade root.

EMISSION MODEL

The calculation domain that would be used by the emission model to predict the emission output of the example combustor is illustrated in Figure 24. Starting with the initial profiles near the dome, the program marches using the recirculation zone as the inner boundary. The mass and specie concentrations in the recirculation zone can be estimated or calculated by several methods, and this provides an entrainment rate and boundary conditions. Once the centerline of the combustor is reached, the inner boundary switches to an axis of symmetry. Cooling slots and radial-injection points can be handled without stopping the marching process. To input the information necessary to perform the above calculation would be quite lengthy and illustrates nothing about the emission program. Usually, the boundary along

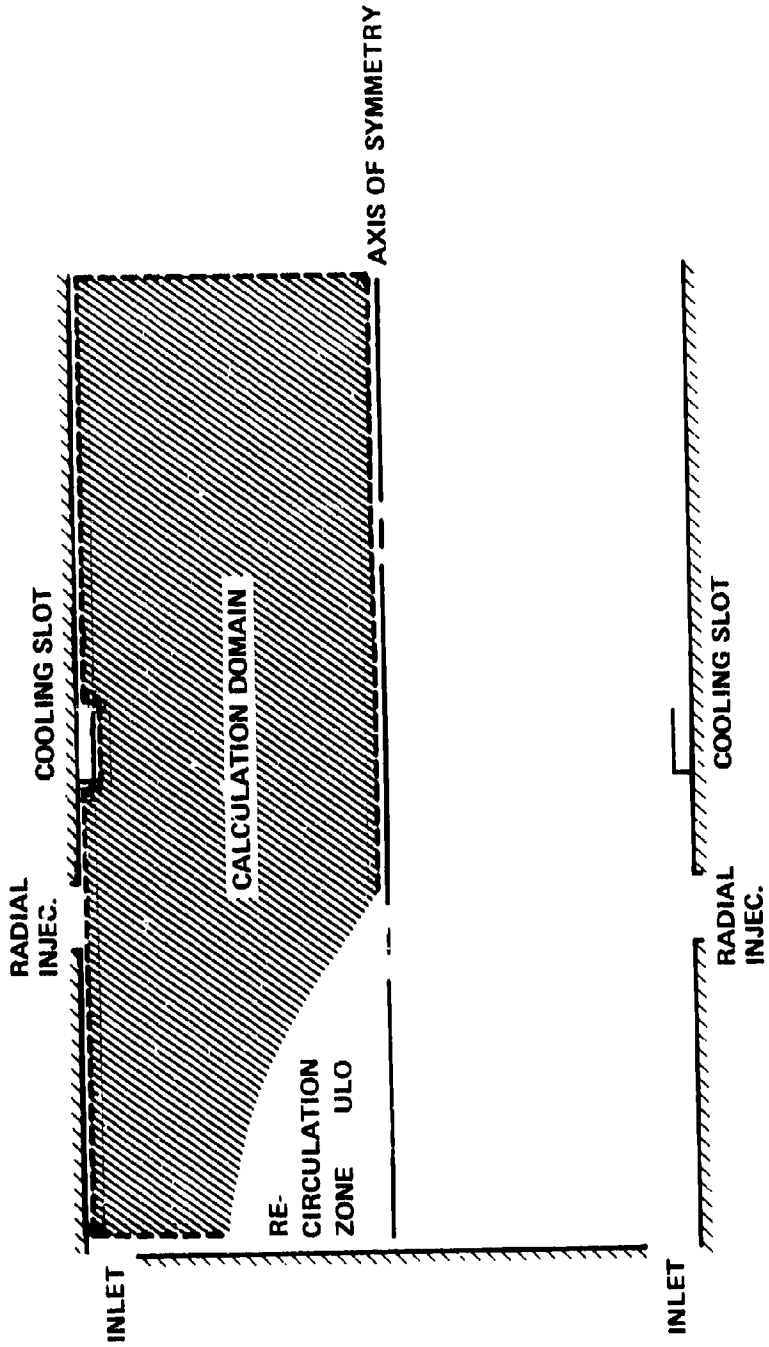


Figure 24. Calculation Domain for Predicting Emission Output of Example Combustor.

the recirculation zone must be handled in a manner that requires internal modifications to the program similar to the Cards MA.528 through MA.743 that are unique to each case analyzed. Therefore, for the example case (Figure 25) the program is started at the downstream edge of the cooling-slot lip and marched to the exit of the liner.

As illustrated in Figure 25, card set 1 is available for case identification. Card 2 specifies 30 cross-stream intervals, axisymmetric geometry, species, two-equation turbulence model, reaction and nonisothermal conditions, while Card 3 produces a profile printout at every 40 marching steps. The initial and final x locations are read on Card 4 and have units of meters, since all the units of the emission model are S.I. The step size was chosen to be small as is necessary with the 16-step kinetic scheme. Since the boundaries are always walls, XOUT AND XEND are set to a large number. A value of POWER equal to 1.0 provides an initial grid that is uniform in y. POWER >1.0 places more nodes near the inner boundary while POWER <-1.0 places more nodes near the outer boundary. Since both walls are straight, only two boundary points are needed, Cards 5 and 6. Cards 7 and 8 define the initial grid, inner and outer radii, and the name-list quantities. Cards 9 and 10 deal with quantities at free boundaries, which are not used, so the data is included only for completeness. The initial profiles are the same as those used in the wall-cooling model example. Nineteen points are used and correspond to the 19 radial nodes used in the 3-D performance model. Three additional profiles, listed in Table 4, are needed for the 16-step-reaction scheme. Since for this case there are no cooling slots or radial injections, the remaining Cards, 24 through 51, are omitted.

CASE TITLE CARD (12A6)

1	EMISSION	MODEL	EXAMPLE				
---	----------	-------	---------	--	--	--	--

CASE TITLE CARD (12A6)

KPLANE	=	7					
--------	---	---	--	--	--	--	--

(8(I2, 8X))

	N	KRAD	MASSTR	ISWRL	MODEL	INERT	IFLUX	ITEMP
2	30	01	01	00	02	02	00	01

	NSTAT	NPROF	NPLOT	ITEST	LATEST			
3	99999	00040	99999		99999			

(5(I5, 5X))

	XU	XULAST	FRA	XEND	XOUT	PRESS	POWER	
4	.03937	.073	.01	10.	10.	101325	1.0	

(8E10.4)

	NBP						
5	02						

(I2)

	X ₁	RI ₁	RE ₁	X ₂	RI ₂	RE ₂	X ₃	RI ₃
6	0	.254	.29972	10.	.254	.29972		

	RE ₃	X ₄	RI ₄	RE ₄	X ₅	RI ₅	RE ...
							(8E10.4)

	RA	RB	RD				
7	0	.254	.29972				

(8E10.4)

Figure 25. Emissions Model Input Sheet (1 of 4)

NAMELIST

8

SINPUT	KREAD=1	PREXD1=2	PEH16	ARGON1=54000.	\$
--------	---------	----------	-------	---------------	----

 CRI=3.0, CX=10, MY=19.28, ERO=1.5, EFU=.5, EO2=1.0
 EH2O=0.

9

UA	UD	VTA	VTD	(8E10.4)			
0	0	0	0				

10

F2A	F2D	TA	TD	T _{wall}	(8E10.4)		
0	0	589.	589.	589.			

11

NIN	NUI	NUE	NVI	NVE	(5 (I2, 8X))		
19	00	00	00	00			

12

Y VALUES								(8E10.4)

13

U VALUES								(8E10.4)

14

Vθ VALUES								(8E10.4)

15

TEMPERATURE VALUES								(8E10.4)

16

MFU VALUES								(8E10.4)

17

MCO ₂ VALUES								(8E10.4)

Figure 25. Emissions Model Input Sheet (2 of 4)

18	MCO VALUES	(8E10.4)							
19	MOX VALUES	(8E10.4)							
20	MH ₂ O VALUES	(8E10.4)							
21	MH ₂ VALUES	(8E10.4)							
22	TURBULENT KINETIC ENERGY	(8E10.4)							
23	TURBULENCE LENGTH SCALE	(8E10.4)							
<p>———— SKIP FOLLOWING CARD SET IF NUI = 0 ————</p>									
24	X - LOC. OF INTERNAL COOLING SLOTS	(8E10.4)							
25	LIP LENGTH OF INTERNAL COOLING SLOTS	(8E10.4)							
26	U - VELOCITY OF INTERNAL COOLING SLOTS	(8E10.4)							
27	VT - VELOCITY OF INTERNAL COOLING SLOTS	(8E10.4)							

Figure 25. Emissions Model Input Sheet (3 of 4)

	TEMPERATURE OF INTERNAL COOLING SLOTS							(8E10.4)
29	FLOW RATE OF INTERNAL COOLING SLOTS							(8E10.4)
30	SLOT HEIGHT OF INTERNAL COOLING SLOTS							(8E10.4)
31	SLOT TO METERING AREA RATIO FOR INT SLOTS							(8E10.4)
SKIP FOLLOWING CARD SET IF NUE = 0								
32	X-LOC OF EXTERNAL COOLING SLOTS							(8E10.4)
33	LIP LENGTH OF EXTERNAL COOLING SLOTS							(8E10.4)
34	U - VELOCITY OF EXTERNAL COOLING SLOTS							(8E10.4)
35	V_T - VELOCITY OF EXTERNAL COOLING SLOTS							(8E10.4)

Figure 25. Emissions Model Input Sheet (4 of 4)

TABLE 4. ADDITIONAL PROFILES FOR EMISSION MODEL.

Node	M_{CO_2}	M_{OX}	M_{H_2O}
19	0	0.232	0
18	0	0.232	0
17	0.0318	0.170	0.0243
16	0.0368	0.163	0.0272
15	0.0311	0.157	0.0291
14	0.0484	0.153	0.0304
13	0.0534	0.151	0.0309
12	0.0494	0.157	0.0282
11	0.0274	0.188	0.0169
10	0.0199	0.200	0.0123
9	0.0527	0.156	0.0282
8	0.0657	0.140	0.0339
7	0.0596	0.142	0.0338
6	0.0537	0.146	0.0333
5	0.0486	0.149	0.0325
4	0.0438	0.152	0.0314
3	0.0400	0.158	0.0293
2	0	0.232	0
1	0	0.232	0

The output of the emission model is shown in Figures 26 and 27. The initial printout consists of some control parameters, the value omega (the transformed cross-stream variable), and the initial profiles. The printing of the variable arrays starts with the value at the inner boundary and continues outward. Since y is always referenced from the inner boundary, $y(1)$ is always zero; therefore, the radius of the inner boundary is printed instead. Figure 27 shows the profiles at the exit of the combustor, $XU = 0.073$ meter. The profiles of NO show small values (much less, however, than one would expect as the principal area of NO formation) occur upstream of where the emission model is started.

FUEL-INSERTION MODEL

The fuel-insertion model could have been used to predict the droplet trajectories of the nozzle in the 3-D performance model example case, Figure 28; however, since that program contains its own spray model, a simple illustrative example was selected instead. A two-dimensional grid 5.0 X 1.5 inches, with the spray originating in the lower left-hand corner and processing a non-uniform flow field, was analyzed.

Card 1 (Figure 28, Sheet 1) allows for case identification. Card 2 specifies the atomizer type, in this case an airblast nozzle with a 10 flow no. and 90-degree cone angle. The other fields are left blank as they pertain to dual orifice and/or simplex nozzles. Cards 3 and 4 are omitted as they are used to input the ΔP versus fuel-flow curve for the secondaries of a dual-orifice nozzle. Card 5 specifies some dimensions of the airblast nozzle, the filming diameter, and exit-flow area along with the airflow rate and temperature. The fuel type, flow rate, and temperature are given on Card 6, plus the flag to specify a nonuniform 2-D field. Other quantities on Card 6 are required only for other

1 TITLE 80
FUEL INSERTION MODEL EXAMPLE

	PRIM FLOW	SEC. FLOW	CONE ANGLE	AIR ASSIST SHROUD EFF AREA	PRIM ORIFICE DIA, IN	SEC ORIFICE DIA, IN		
	NO.	NO.	DEG	IN ²	IN	IN		
	PPH/ $\sqrt{\text{PSI}}$							
	1	2	3	4	5	6	7	8
2	3	110	90					

*ATOMIZER TYPE: 00001 = SIMPLEX
 00002 = DUAL ORF
 00003 = AIR BLAST

**AIR ASSIST: 00001 = NO ASSIST
 00002 = WITH ASSIST

*FOR ATOM TYPE = 00002 (DUAL ORIF) ONLY (LEAVE OUT FOR OTHERS)
 INPUT SECONDARY FLOW SCHEDULE

W_s = SECONDARY FUEL FLOW, LB/HR

ΔP_s = $\Delta P_{\text{SEC. ORIF}}$ + $\Delta P_{\text{FLOW DIVIDER VALVE}}$ P/D

CRACK
POINT
FLOW

	W_{s1}	W_{s2}	W_{s3}	W_{s4}	W_{s5}	
	1	11	21	31	41	51
3						

CRACK
PRESSURE

	ΔP_1	ΔP_2	ΔP_3	ΔP_4	ΔP_5	
	1	11	21	31	41	51
4						

*FOR ATOM TYPE = 00003 (AIRBLAST) ONLY (LEAVE OUT FOR OTHERS)

	FILMING DIA IN.	NOZZLE AIRFLOW LB/SEC	FLOW AREA IN ²	AIR TEMP °R
	1	11	21	31
5	0.5	.05	.2	1060

Figure 28. Fuel Insertion Model Input (1 of 3)

 *** AIR FUEL FUEL FUEL AIR AIR
 FUEL FLOW TEMP FLOW ΔP ΔP ASSIST ASSIST
 TYPE OPTION °R LB/HR PSI PSI °R
 1 - 6 - 11 21 31 41 51 61 71

6	2	2	520.	60.					
---	---	---	------	-----	--	--	--	--	--

FUEL | 00002=JP5 | *AIR |
 TYPE | 00004=JP4 | FLOW | 00001=UNIFORM GAS STREAM
 OPTION | | | 00002=2-D FIELD OPTION

T_{GAS}, °R V_{GAS}, P_{GAS}, X_{MAX}, Y_{MAX}, ← UNIFORM STREAM
 1 11 FPS 21 PSIA 31 IN. 4 °R 51 OPTION

7	.2	.2	147.	5.0	1.5			
---	----	----	------	-----	-----	--	--	--

X_{NOZ} IN. Y_{NOZ} IN. P_{GAS} PSIA X_{MAX} IN. Y_{MAX} IN. ← Z-D FIELD OPTION

CARDS 8 THROUGH 13 SKIPPED IF AIR FLOW OPTION = 00001

IN = NO. OF X-DIR POINTS IN 2-D FIELD
 JN = NO. OF Y-DIR POINTS

8	03	03						
---	----	----	--	--	--	--	--	--

X VALUES (SE10.4) X - LOCATIONS OF GRID POINTS (FT)

9	0.	.2083	.4167					
---	----	-------	-------	--	--	--	--	--

Y VALUES (SE10.4) Y - LOCATIONS OF GRID POINTS (FT)

10	0.	.0625	.125					
----	----	-------	------	--	--	--	--	--

READ ALONG + X LINES STARTING WITH
 U VALUES (SE10.4) SMALLEST Y VALUE (FT/SEC)

11	30.	75.	100.					
	35.	80.	110.					
	40.	85.	115.					

Figure 28. Fuel Insertion Model Input (2 of 3)

READ ALONG + X LINES STARTING WITH
SMALLEST Y VALUE (FT/SEC)

V VALUES (SE10.4)

12	0.	0.	0.				
	0.	0.	0.				
	0.	0.	0.				

READ ALONG + X LINES STARTING WITH
SMALLEST Y VALUE (DEG. R)

T VALUES (SE10.4)

13	1060	1060	1060				
	1060	1060	1060				
	1060	1060	1060				

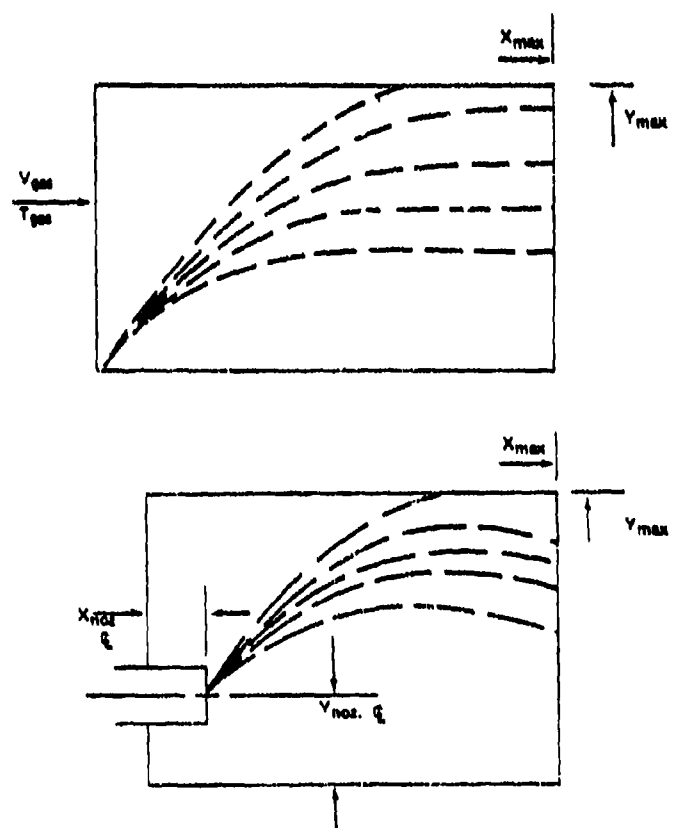


Figure 28. Fuel Insertion Model Input (3 of 3)

nozzle types. The values of Card 7 can have two meanings, depending on the type of flow field. In this case, they give the X and y location of the origin of the spray, the pressure, and limits of the two-dimensional grid. Cards 8 through 13 are required in this example, and it can be seen that only a 3 X 3 grid was used which requires 9 values of U, V, and temperature.

The program output is illustrated in Figures 29 and 30. The first items printed are some of the input quantities, along with the specific gravity and viscosity of the specified fuel. The fuel ΔP and velocity are given next, along with the calculated value of SMD. For each of the five droplets, the locations are then given at the point that selected fractions of the fuel that had evaporated. In this case, droplets 1 and 2 evaporated within the specified boundaries. Figure 30 shows the trajectory output for each of the five droplets. The output consists of a pair of lines, the first giving the X and Y location of the droplet and the second giving the diameter, temperature, velocity, and fraction evaporated.


```

FUEL INSERTION MODEL EXAMPLE
AIRCRAFT ALTITUDE, COME ANGLE = 90.0
WITHOUT AIR ASSIST          -5000
FILM TH. NOZZLE AIRFLOW, FLOW AREA, 135, 2, 5000
FUEL TQPT TPUEL PPH DELT DELPSH TSM = 2, 420.00000 40.00000 -0.00000 -0.00000
ARBITRARY FLOW OPTION, X START, Y STRAT, P GAS, Y MAX, Y MIN = .20000 -2.00000 147.00000 5.00000 1.50000
SG = .0260E+00 (DIM S) VISC FUEL = .2492E+01 CENTISTIMES

DELTA P (1) = 30.80 PPM (1) = 60.80
DELTA P (2) = 0.00 PPM (2) = 0.00
DELTA P (3) = 0.00 VFUEL = 74.39
SMD = 41.20

AIR/FUEL MASS RATIO = 3.000 EFF. AIR VELOCITY =
DROPLET 1 E = .10 X1, X2 = .43062 .36982 IM
DROPLET 1 E = .20 X1, X2 = .52314 .40377 IM
DROPLET 1 E = .30 X1, X2 = .50328 .42652 IM
DROPLET 1 E = .40 X1, X2 = .68542 .44966 IM
DROPLET 1 E = .50 X1, X2 = .77486 .45777 IM
DROPLET 1 E = .60 X1, X2 = .87766 .47021 IM
DROPLET 1 E = .70 X1, X2 = 1.00161 .48158 IM
DROPLET 1 E = .80 X1, X2 = 1.16126 .49217 IM
DROPLET 1 E = .90 X1, X2 = 1.38888 .50199 IM
DROPLET 1 E = .95 X1, X2 = 1.56262 .50668 IM
DROPLET 1 HAS EVAPORATED

DROPLET 2 E = .10 X1, X2 = .65474 .49756 IM
DROPLET 2 E = .20 X1, X2 = .84873 .56477 IM
DROPLET 2 E = .30 X1, X2 = 1.01970 .60716 IM
DROPLET 2 E = .40 X1, X2 = 1.19738 .63969 IM
DROPLET 2 E = .50 X1, X2 = 1.39648 .66632 IM
DROPLET 2 E = .60 X1, X2 = 1.61664 .68880 IM
DROPLET 2 E = .70 X1, X2 = 1.84389 .70746 IM
DROPLET 2 E = .80 X1, X2 = 2.06680 .72239 IM
DROPLET 2 E = .90 X1, X2 = 2.02991 .73325 IM
DROPLET 2 E = .95 X1, X2 = 3.50925 .73773 IM
DROPLET 2 HAS EVAPORATED

DROPLET 3 E = .10 X1, X2 = .86633 .61133 IM

```

Figure 29. Fuel Insertion Model Output.

8 - 20 / VOLUME		20 - 40 / VOLUME		40 - 60 / VOLUME		60 - 80 / VOLUME		80 - 100 / VOLUME	
25.647		38.397		55.615		59.737		78.277 MICRONS	
I	Vf E+100 /	I	Vf E+100 /	I	Vf E+100 /	I	Vf E+100 /	I	Vf E+100 /
.20	.20	.20	.20	.21	.20	.20	.20	.20	.20
26. 520.	74. 0. /	38. 520.	74. 0. /	49. 520.	74. 0. /	60. 520.	74. 0. /	78. 520.	74. 0. /
.24	.23	.24	.23	.24	.23	.31	.30	.31	.30
26. 576.	62. 0. /	39. 547.	64. 0. /	49. 576.	64. 0. /	60. 561.	57. 0. /	79. 547.	56. 0. /
.27	.27	.27	.27	.27	.27	.39	.36	.39	.37
26. 612.	54. 1. /	39. 573.	57. 3. /	49. 557.	60. 0. /	60. 597.	51. 0. /	79. 564.	54. 0. /
.31	.30	.31	.30	.21	.30	.47	.42	.47	.43
26. 559.	48. 2. /	39. 599.	52. 0. /	49. 574.	55. 0. /	61. 612.	47. 1. /	79. 582.	50. 0. /
.36	.33	.36	.33	.35	.33	.55	.47	.55	.48
26. 607.	45. 4. /	39. 623.	49. 33. /	49. 593.	51. 33. /	61. 635.	45. 1. /	79. 598.	47. 0. /
.40	.35	.40	.35	.40	.36	.64	.52	.64	.54
26. 729.	42. 7. /	39. 646.	47. 1. /	49. 610.	49. 1. /	61. 658.	44. 2. /	79. 616.	46. 1. /
.44	.37	.44	.37	.43	.39	.73	.57	.73	.58
26. 756.	40. 11. /	39. 667.	45. 2. /	49. 627.	47. 1. /	61. 676.	44. 3. /	79. 629.	45. 1. /
.49	.39	.49	.39	.47	.42	.82	.61	.82	.63
26. 779.	40. 16. /	39. 687.	44. 3. /	49. 643.	45. 1. /	61. 696.	44. 4. /	80. 643.	45. 1. /
.54	.41	.52	.44	.52	.44	.92	.55	.92	.60
26. 797.	40. 22. /	39. 705.	43. 5. /	49. 657.	44. 2. /	61. 710.	44. 5. /	80. 657.	45. 2. /
.58	.42	.57	.46	.56	.47	1.01	.68	1.01	.69
26. 812.	40. 28. /	39. 721.	42. 6. /	49. 671.	44. 2. /	60. 724.	44. 7. /	80. 669.	46. 2. /
.63	.43	.61	.48	.60	.49	1.10	.71	1.09	.75
26. 822.	41. 34. /	39. 734.	42. 8. /	49. 684.	43. 3. /	60. 737.	44. 8. /	80. 681.	46. 3. /
.68	.44	.66	.50	.65	.51	1.20	.76	1.18	.78
23. 831.	42. 40. /	39. 750.	42. 10. /	49. 697.	43. 5. /	60. 749.	44. 10. /	80. 691.	47. 4. /
.73	.45	.70	.52	.69	.53	1.30	.77	1.28	.81
22. 838.	43. 45. /	38. 762.	42. 12. /	49. 708.	43. 5. /	60. 760.	44. 12. /	80. 701.	47. 4. /
.78	.45	.75	.53	.74	.55	1.39	.78	1.37	.84
21. 844.	44. 51. /	38. 773.	42. 15. /	49. 719.	43. 6. /	59. 770.	50. 14. /	79. 710.	50. 5. /
.83	.46	.80	.55	.79	.57	1.44	.81	1.47	.87
21. 849.	45. 56. /	38. 783.	43. 17. /	49. 729.	43. 7. /	59. 778.	52. 18. /	79. 719.	51. 6. /
.88	.47	.85	.56	.83	.59	1.59	.83	1.58	.89
20. 853.	46. 60. /	37. 792.	44. 20. /	49. 739.	43. 9. /	59. 786.	54. 18. /	79. 727.	53. 7. /
.93	.48	.89	.58	.88	.61	1.69	.85	1.66	.91
19. 857.	47. 65. /	37. 800.	44. 23. /	49. 748.	44. 10. /	58. 793.	55. 20. /	79. 734.	54. 8. /
.98	.48	.94	.59	.93	.63	1.79	.87	1.76	.94
18. 860.	47. 68. /	37. 804.	45. 24. /	49. 756.	44. 11. /	58. 799.	57. 23. /	79. 741.	54. 9. /
1.03	.49	.99	.60	.97	.64	1.82	.89	1.80	.94
18. 864.	48. 72. /	36. 812.	46. 28. /	48. 764.	45. 13. /	57. 805.	58. 25. /	79. 748.	57. 10. /
1.08	.49	1.04	.61	1.02	.65	1.96	.90	1.94	.97
17. 867.	49. 75. /	36. 818.	47. 31. /	48. 771.	45. 14. /	56. 810.	60. 27. /	78. 754.	58. 11. /
1.13	.49	1.09	.62	1.07	.67	2.06	.91	2.05	.99
18. 870.	50. 78. /	35. 823.	48. 34. /	48. 778.	46. 16. /	56. 815.	62. 30. /	78. 760.	60. 12. /
1.18	.49	1.14	.63	1.12	.66	2.18	.93	2.15	1.01
18. 873.	51. 81. /	35. 827.	49. 37. /	48. 784.	47. 17. /	55. 819.	63. 32. /	78. 765.	62. 13. /
1.23	.50	1.19	.65	1.17	.70	2.28	.94	2.25	1.02
15. 875.	52. 83. /	34. 831.	50. 40. /	47. 790.	47. 19. /	55. 823.	65. 34. /	78. 770.	63. 14. /
1.28	.50	1.24	.65	1.22	.71	2.36	.95	2.35	1.04
14. 878.	53. 86. /	34. 834.	51. 42. /	47. 795.	48. 21. /	54. 828.	67. 36. /	77. 775.	65. 15. /

Figure 30. Fuel Insertion Model Output.

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APPENDIX A
INPUT SHEETS

ANNULUS LOSS MODEL INPUT SHEET

										ITER	IPIC	IDBUG				
										78	79	80				
1	TITLE - MUST HAVE FOR EACH CASE															
1	ANNULUS INLET FLOW CARD															
	NELNELI															
	1	345678	11	W1	21	PT1	31	TT1	41	BETA1	51	DP/P	61	71		
2	<input checked="" type="checkbox"/>	<input checked="" type="checkbox"/>														
	DOMES INLET FLOW CARD (REQUIRED FOR INTERNAL LINER FLOW ONLY)															
	NELI															
	12	678	11	W	21	PT	31	TT	41	BETA	51	61	71			
3	<input checked="" type="checkbox"/>	<input checked="" type="checkbox"/>														
	CONSTANTS CARD															
	1		11	FRIC	21	BLKF	31	TANS	41	CDB	51	SK	61	RG	71	IREAX
4	<input checked="" type="checkbox"/>															

- 1 Title - Run ident appears on printed output and plot
- ITER {
 = 1 some (or all) holes fixed, inlet flow fixed, iterate to get pressure drop.
 = 2 some (or all) holes fixed, pressure drop fixed iterate to get inlet flow (input W1 is first guess).
- IPIC {
 = 0 no plot
 = 1 plot drawn
- IDBUG {
 = 0 output printed after converged solution
 = 1 output printed after each iteration } Use only to de-
 = 2 output printed after each element } bugg
 non-convergence
- 2 3 NEL = total number of element stations (card 2 only)
 NELI = element ID no. (NELEM) at annulus or dome inlet 2 and 3
- W₁ and W = air flow at inlet stations, lb/sec
 PT₁ and PT = inlet total pressure, PSIA
 TT₁ and TT = inlet total temperature, R
 Beta₁, Beta = swirl angle at inlet
 DP/P = total pressure drop, PSIA/PSIA (first guess if ITER = 1) card 2 only
- 4 FRIC {
 = 0, smooth wall friction factor
 = -1, no wall friction
 = roughness factor for rough walls
 BLKF = annulus effective area factor (= .83 for fully developed turbine flow)
 TANS = tangent of flow separation spread angle (.1 recommended)

CDB = drag coefficient of struts across annulus (1-1.2 RECM)
SK = ratio of air specific heats,
RG = air gas constant
IREAX } = 0. for reverse flow annular or can combustors
 } = 1. for axial flow annular. First data set is for OD
 } panel. Program expects a second set for ID panel

CASE TERMINATION

After last card of case:

- o In Column 1, Column 2 blank - case repeated with changes, next card is title card followed by cards with changes from previous run.
- oo In Columns 1 and 2, next card is EOF to quit or new title card followed by all cards for complete new case.

SELEM

L		NP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L		SP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L		SP	RP	XL	RL	CL	T RISE		
F		W/W1	N HOLES	NHTYP	CD		I SEP		
F		W/W2	N HOLES	NHTYP	CD	D HOLES	I SEP		
F		W/W1					I SEP		
F		W/W1					I SEP		
F		W/W1					I SEP		

	1	2	145	11	11	11	41	51	61	71	76
1											
2											
3											
4											
5											
6											
7											
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38											
39											
40											

PROGRAM 117 INPUT DATA SHEET
INPUT FORMAT FOR ELEMENT CARDS - SHEET 1

NELEM		ALL NUMBERS MUST HAVE DECIMAL POINTS										
1	2	3	4	5	11	21	31	41	51	61	71	76
L	0	0	1		XP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L	I	0	0	2	XP	RP	XL	RL	CL	T LINER	BLK I	DBK I
L	C	0	0	3	SP	RP	XL	RL	CL	T RISE	--	--
F	F	0	0	4	W/WI	N HOLES	NHTYP	CD	--	I SEP	--	--
F	D	0	0	5	W/WI	N HOLES	NHTYP	CD	D HOLES	I SEP	--	--
F	B	0	1	0	W/WI	--	--	--	--	I SEP	--	--
F	I	0	2	0	W/INJ/W	V JET	ANGJ	ABETA	--	I SEP	--	--

ELEMENT SPECIFICATION

Flow passage is divided into length (L) and flow (F) elements. element numbers, NELEM, can be in arbitrary order, i.e., 10, 1, 3, 4, 16, 30. The cards are stacked in order from inlet to last P because numbers are arbitrary, a new element can be inserted without renumbering other cards.

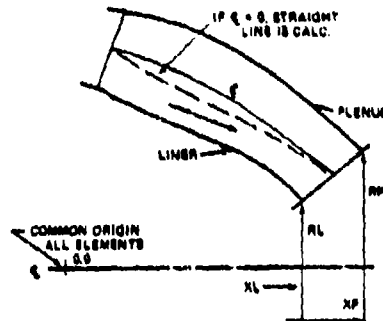
L. LENGTH ELEMENTS All Dimensions in inches

First L card is annulus inlet (CL = 0) - For both external and internal cases. Internal and external flow cases must be run separately

For internal cases, 2nd card is dome inlet (LI) and LC cards are used with T RISE = AT due to combustion in this element (no L cards)

For external cases use only L Cards

- XL, XP = X COORD to end of element L = Liner
- RL, RP = Radius to end of element P = Plenum
- (For internal flow XP, RP = OD, XL, RL = ID)
- CL = Length of element (optional)
- TLIN = Mean wall temp. over CL, °R
- If = 0 then TLIN = TTI
- BLKI = Frontal Area of Struts
- Annulus Area
- DBKI = Width of strut



F. ORIFICE FLOW ELEMENT

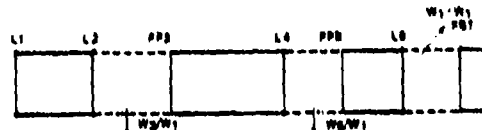
F Cards are inserted between L cards at points where flow is extracted. Flow conditions into F elem. are those from upstream L elem.

- Types: FF = Fixed Flow Ratio, W/WI
- FD = Fixed Orf. Diam. (W/WI is First Guess)
- FB = Bleed flow (not included in liner flow)
- FI = Internal flow elem. (input to these elements is obtained from an external flow solution)

- W/WI = Orifice Flow/Inlet Flow
- NHOLES = No. of Orifices
- NHTYP = Hole Type (For CD)
 1. Flush Hole, Thin Wall
 2. Plunged Hole
 3. Cooling Skirt
 4. Flush Hole, Thick Wall
 5. CD Input
 6. Rectangular Hole

- CD = DISCH Coefficient (NHTYP = 5 Only)
- DHOLES = Hole Diam. (for FD Only)
- I SEP = 1, Separation is Reattached
- VJET = Jet Injection Velocity, FPS
- ANGJ = Jet Injection Angle, Deg.
- ABETA = Swirl Angle in Annulus Outside FI Elem.

FLOW ELEMENTS



3D COMBUSTOR PERFORMANCE MODEL

22 VARIABLE TITLE CARDS (4A10) SAME ORDER AS IMPRINT

1

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CASE TITLE CARD (8A10)

2

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LPI MPl LPI IPLAX MODEL MODER IPAR ITRAD

3

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IU MODEN INTAPE IDW IRES

4

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ISOLVE (8 (I2, 8X))

U V W P' KE e D MFU

5

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MCO R FX FY FZ

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ICTDMA (8 (I2, 8X))

U V W P' KE e D MFU

6

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MCO R

--	--	--	--	--	--	--	--

IMPRINT (8 (I2, 8X))

U V W PRESS KE f.m D MFU

7

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TEMP	\bar{n}	Favg	Fx	Fy	Fz	MCO	MH2O

MO2	MCO2	MN2	μ_{EFF}	DENSITY	EVAP		

RELAXATION PARAMETERS (8E10.4)

U	V	W	P'	KE	ϵ	D	MFU

MCO	\bar{n}	Fx	Fy	Fz	PRESS	DENSITY	VISCOS

LAMINAR PRANDTL NUMBERS (8E10.4)

U	V	W	P'	KE	ϵ	D	MFU

MCO	\bar{n}						

TURBULENT PRANDTL NUMBERS (8E10.4)

U	V	W	P'	KE	ϵ	D	MFU

MCO	\bar{n}						

X-COORDINATES (1-LP1) (8E10.4)

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12 RI Y-COORDINATES (2-MP1) (8E10.4)

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13 Z-COORDINATES (1-NP1) (8E10.4)

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14 IWEI JWIO (2(I2,8X))

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15 IWLI VALUES (8(I2,8X)) (SKIP 15 AND 16 IF IWEI = 2)

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JWLO VALUES (8 (12, 8X))

16

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PRESS DEN ABSOR SCATR AKFAC ALFAC (8E10.4)

17

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CX HY HFU FUMCO (8E10.4)

18

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PREXP1 ARCON1 CR1 PREXP2 ARCON2 CR2 (8E10.4)

19

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(8E10.4)

C1 C2 CD AMU ERROR TCYLW TINLW TLIP

20

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(2 (13, 7X), 6 (12, 8X))

LASTEP IJUMP JSW1 JSW2 NUINJ NVINJ

21

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USW VSW SWNO AFSW FSW TSW (8E10.4)

22

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NFNZ ISPRAY TFUEL

23

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XO YO ZO ALFA BETA DELTA THETA 1 THETA 2

24

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RSP WF SMD VFUEL (SKIP 24 IF NFNZ = 00)

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(SKIP CARDS 25 → 30 IF NUINJ = 00)

I - LOCATION OF COOLING SLOTS (8(I2, 8X))

25								
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J - LOCATION OF COOLING SLOTS (8(I2, 8X))

26								
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AXIAL SLOT VELOCITY (8E10.4)

27								
----	--	--	--	--	--	--	--	--

TANG. SLOT VELOCITY (8E10.4)

28								
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SLOT FLOW RATE (8E10.4)

29								
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SLOT TEMPERATURE (8E10.4)

30								
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(SKIP CARDS 31 → 38 IF NVINJ = 00)

I - LOCATION OF RADIAL INJECTION (8(I2, 8X))

31								
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J - LOCATION OF RADIAL INJECTION (8(I2, 8X))

32								
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K - LOCATION OF RADIAL INJECTION (8(I2, 8X))

33								
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INJECTION VELOCITY (8E10.4)

34								
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INJECTION TURBULENT KINETIC ENERGY (8E10.4)

35

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INJECTION TURBULENT LENGTH SCALE (8E10.4)

36

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INJECTION FLOW RATE (8E10.4)

37

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INJECTION TEMPERATURE (8E10.4)

38

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3D COMBUSTOR PERFORMANCE MODEL
INPUT SHEET DESCRIPTION

Card Set	Description
1	Each card is a heading for a particular three-dimensional array that is printed out. These never change.
2	Case title card
3	LPI - Number of grid nodes in axial (x) dir. MPI - Number of grid nodes in radial (y) dir. NPI - Number of grid nodes in tang. (z) dir. IPLAX - 01 For plane geometry - 02 For axisymmetric geometry MODEL - 01 For laminar viscosity - 02 For K-E viscosity model MODER - 01 For kinetic controlled combustion - 02 For kinetic and turbulence controlled combustion IPAR - 01 For absolute pressure - 02 For relative pressure ITRAD - 01 No radiation - 02 With radiation
4	IU - 01 Input units are international system (i.e., meters, kilograms, degrees kelvin, newtons, joules, radians, seconds or combinations thereof) - 02 User selected input units MODEN - 01 Density is fixed at the value of "Den" on Card 17 - 02 Density calculated from perfect gas law INTAPE - 00 Initial conditions not printed - 08 Initial conditions printed IDW - 00 Inner boundary is axis of symmetry - 01 Inner boundary is wall IRES - 00 This is a new case - 01 This is a restart of previous case
5	An 01 in proper field indicates that this particular variable will be solved for; an 00 indicates that it will not be.
6	Indicates the number of "sweeps" made in the solve routine for each variable.

- 7 An 01 indicates that this variable will be printed,
an 00 indicates that it will not be.
- 8 Relaxation parameters for each variable.
- 9 Laminar Prandtl numbers for each variable.
- 10 Turbulent Prandtl numbers for each variable.
- 11 X-coordinates (LPI values)
- 12 RI - Radius of inner boundary
Y-coordinates as measured from inner
boundary (MPI-1) values. Since Y(1) is
always 00, RI is read in its place.
- 13 Z - coordinates (NPI values)
- 14 IWEI - I-node at which inclined wall ends
JWIO - J-node at which inclined wall starts
- 15 IWLI - Starting I-nodes of the calculation domain
when inclined wall is present
- 16 JWLO - Ending J-nodes of the calculation domain
when inclined wall is present
- 17 PRESS - System pressure
DEN - The value of density if option MODEN = 01
is selected
ABSOR - Absorption coefficient in radiation model
SCATR - Scattering coefficient in radiation model
AKFAC - Internally defined turbulent kinetic
energies are AKFAC times the appropriate
velocity squared
ALFAC - Internally defined turbulent length scales
are ALFAC times the appropriate distance
- 18 CX - Carbon atoms in fuel molecule
HY - Hydrogen atoms in fuel molecule
HFU - Heat of formation of fuel
FUMCO - Initial value assigned to M_{CO}
- 19 PREXPI - Preexponent of 1st reaction
ARCONI - Activation energy divided by gas constant
of 1st reaction (E/R)
CR1 - Constant in turbulence controlled reaction
rate for 1st reaction
PREXP2 - Preexponent of 2nd reaction

- ARCON2 - Activation energy divided by gas constant
of 2nd reaction (E/R)
- CR2 - Constant in turbulence controlled reaction
rate for 2nd reaction
- 20 C1 - Turbulence model constant
C2 - Turbulence model constant
CD - Turbulence model constant
AMU - The value of the viscosity if option model
= 01 is specified. Also the laminar
viscosity used in the "wall functions"
- ERROR - Program will terminate if total error in
mass becomes less than this value
- TCYLW - Temperature of cylindrical portion of
combustor wall
- TINLW - Temperature of inclined wall portion of
combustor and of dome.
- TLIP - Temperature of cooling slot lip.
- 21 LASTEP - Maximum number of iterations
IJUMP - Number of iterations between array
printout
- JSW1 - J-node at start of dome inlet
JSW2 - J-node at end of dome inlet
NUINJ - Number of axial injection points (cooling
slots)
NUINJ - Number of radial injection points
- 22 USW - Axial velocity of dome inlet
VSW - Radial velocity of dome inlet
SWNO - Ratio of tangential to axial velocity at
dome inlet
- AFSW - Flow rate of fuel and air through dome inlet
FSW - Flow rate of fuel through dome inlet
TSW - Temperature at dome inlet
- 23 NFNZ - 00 No liquid fuel nozzle
- 01 Liquid fuel nozzle present
ISPRAY - Droplet evaporation routine is called
every ISPRAY iterations
TFUEL - Initial temperature of liquid fuel
- 24 XO - X-location of origin of fuel nozzle spray
YO - Y-location of origin of fuel nozzle spray
ZO - Z-location of origin of fuel nozzle spray
ALFA - Nozzle cone angle
BETA - Nozzle back angle
DELTA - Nozzle down angle
THETA1 - Initial spray cone segment angle
THETA2 - Final spray cone segment angle

RSP - Number of spray cone rays
WF - Fuel flow rate
SMD - Sauter mean diameter
VFUEL - Initial fuel droplet velocity

- 25 I node location of cooling slots
- 26 J node location of cooling slots
- 27 Cooling slot axial velocity
- 28 Cooling slot tangential velocity
- 29 Cooling slot mass flow rate
- 30 Cooling slot temperature
- 31 I node location of radial injection
- 32 J node location of radial injection
- 33 K node location of radial injection
- 34 Radial injection velocity
- 35 Radial injection turbulent kinetic energy
- 36 Radial injection turbulence length scale
- 37 Radial injection mass flow rate
- 38 Radial Injection temperature

LINER COOLING MODEL INPUT SHEET

1 CASE TITLE (12A6)

--	--	--	--	--	--	--	--

CASE TITLE (12A6)

--	--	--	--	--	--	--	--

(8 (12, 8X))

2 N KRAD MASSTR ISWRL MODEL INERT IFLUX ITEMP

--	--	--	--	--	--	--	--

3 NSTAT NPROF NPLOT ITEST LASTEP (5 (15.5X))

--	--	--	--	--	--	--	--

4 XU XULAST FRA XEND XOUT PRESS POWER (8E10.4)

--	--	--	--	--	--	--	--

5 NBP (12)

--	--	--	--	--	--	--	--

(8E10.4)

6 X₁ RI₁ RE₁ X₂ RI₂ RE₂ X₃ RI₃

--	--	--	--	--	--	--	--

RE₃ X₄ RI₄ RE₄ X₅ RI₅ RE₅... (8E10.4)

--	--	--	--	--	--	--	--

7 RA RB RD (8E10.4)

--	--	--	--	--	--	--	--

--	--	--	--	--	--	--	--

NAME LIST

8 INPUT

								\$
--	--	--	--	--	--	--	--	----

9 UA UD (8E10.4)

--	--	--	--	--	--	--	--	--

10 F2A F2D TA TD T_{wall} (8E10.4)

--	--	--	--	--	--	--	--	--

11 NIN (I2)

--	--	--	--	--	--	--	--	--

12 Y VALUES (8E10.4)

--	--	--	--	--	--	--	--	--

13 U VALUES (8E10.4)

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14 (8E10.4)

--	--	--	--	--	--	--	--	--

15 TEMPERATURE VALUES (8E10.4)

--	--	--	--	--	--	--	--	--

16 MFU VALUES (8E10.4)

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17 PHI VALUES (8E10.4)

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MCO VALUES (8E10.4)
18

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TURBULENT KINETIC ENERGY (8E10.4)
19

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TURBULENCE LENGTH SCALE (8E10.4)
20

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VELA_I VELA_O TAN_I TAN_O PR_I PR_O (8E10.4)
21

--	--	--	--	--	--	--	--

(12, 8X, 7E10.4)
NFNZ XDP YDP UF VF SMD WF TFUEL
22

--	--	--	--	--	--	--	--

LINER COOLING MODEL
INPUT SHEET DESCRIPTION

Card Set	Description
1	Case title cards
2	<p>N - Number of cross stream intervals</p> <p>KRAD - 00 For plane geometry 01 For axisymmetric geometry</p> <p>MASSTR - 00 Species equations not solved 01 Species equations solved</p> <p>ISWRL - 00 Always</p> <p>MODEL - 01 Laminar viscosity 02 Two-equation K-E viscosity model</p> <p>INERT - 01 Species are inert 02 Species will react</p> <p>IFLUX - 00 Wall adiabatic 01 Wall temperature calculated</p> <p>ITEMP - 00 Isothermal 01 Enthalpy solved</p>
3	<p>NSTAT - Number of steps between printout of output variables</p> <p>NPROF - Number of steps between printout of output variables and profiles</p> <p>NPLOT - Number of steps between line-printer plots</p> <p>ITEST - 00 No extra printout 01 Extra printout</p> <p>LASTEP - Maximum number of marching steps</p>
4	<p>XU - Initial X-location</p> <p>XULAST - Final X-location</p> <p>FRA - Fraction of grid height to be used as step size</p> <p>XEND - X-location of end of inner wall</p> <p>XOUT - X-location of end of outer wall</p> <p>PRESS - Pressure</p> <p>POWER - Control for node spacing</p>
5	NBP - Number of boundary pairs
6	<p>X_1 - X-Location at which boundary is specified</p> <p>RI_1 - Inner boundary radius</p> <p>RE_1 - Outer boundary radius</p>
7	<p>RA - Radius of axis of symmetry</p> <p>RB - Inner boundary radius at initial X-location</p> <p>RD - Outer boundary radius at initial X-location</p>

- 8 Namelist, see page 158 for variables
- 9 UA - Axial velocity at "free" inner boundary
 UD - Axial velocity at "free" outer boundary
- 10 F2A - Fuel mass fraction at "free" inner boundary
 F2D - Fuel mass fraction at "free" outer boundary
 TA - Temperature at "free" inner boundary
 TD - Temperature at "free" outer boundary
 TWALL - Wall temperature
- 11 NIN - Number of points on input initial profile
- 12 Y-values of input initial profile (NIN values)
- 13 U-values of input initial profile
- 14 Blank card(s)
- 15 Temperature values of input initial profile
- 16 MFU values of input initial profile
- 17 Total fuel values of input initial profile
- 18 M_{CO} values of input initial profile
- 19 KE values of input initial profile
 (Read only if KREAD = 1)
- 20 f_m Values of input initial profile
 (Read only if KREAD = 1)
- 21 VELA_I - Inner annulus velocity
 VELA_O - Outer annulus velocity
 TAN_I - Inner annulus temperature
 TAN_O - Outer annulus temperature
 PR_I - Radius of inner plenum
 PR_O - Radius of outer plenum
- 22 NFNZ - 00 No fuel nozzle
 01 Fuel nozzle present
 XDP - X-location of fuel nozzle
 YDP - Y-location of fuel nozzle
 UF - Axial velocity of fuel droplets
 VF - Radial velocity of fuel droplets
 SMD - Sauter mean diameter
 WF - Fuel mass flow rate
 TFUEL - Fuel temperature

LINER COOLING MODEL
NAMELIST INPUT

VBL	Value	Description
IUTRAP	2	Test for negative U's, see STRIDE(2)
ULIM	0.05	Entrainment control
PEILIM	0.01	Max. fractional mass flow change per step
AFAC	0.2	Relaxation on duct area deviation
AEXDLM	0.02	Max. duct area fractional deviation per step
NOVEL	2	01-U not solved for, 02-solve for U
ARCON1	28500	Activation energy divided by gas constant for fuel reaction
PREXP1	5E + 15	Preexponent for fuel reaction
CR1	6.0	Eddy breakup constant for fuel reaction
ARCON2	12500	Activation energy divided by gas constant for CO reaction
PREXP2	6E + 8	Preexponent for CO reaction
CR2	4.0	Eddy breakup constant for CO reaction
CP	1048	Specific heat
IPDDX	1	01 - Std genmix pressure grad. calculation 02 - "Grid filling duct" version
C1	1.42	Turb. Constant
C2	1.92	Turb. Constant
CD	0.09	Turb. Constant
C2VT	0.36	Turb. Constant
AKFAC	0.03	Factor for internally generated kinetic energy profiles, $KE = AKFAC * U^2$
ALFAC	0.02	Factor for internally generated length scale profiles, $l_m = ALFAC * \Delta Y$
PREF	--	Turbulent Prandtl numbers
CX	1.0	Molecular carbon value of fuel
HY	4.0	Molecular hydrogen value of fuel
HFU	-49317	Heat of formation of fuel
MODER	2	1 - Kinetic only, 02 - Kinetic + Diffusion
ITHIN	1	Thin output profiles by printing every ITHIN value
KREAD	0	0 - K & E profiles generated internally 1 - K & l_m read in
URAT	0.83	Cooling slot velocity shape factor
ABSOR	0.1	Absorption coefficient in radiation model
EMISW	0.8	Emissivity of liner wall

TRANSITION MIXING MODEL INPUT SHEET

1 CASE TITLE CARD (12A6)

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CASE TITLE CARD (12A6)

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2 (8 (12, 8X))

N	KRAD	MASSTR	ISWRL	MODEL	INERT	IFLUX	ITEMP

3 (5 (15, 5X))

NSTAT	NPROF	NPLOT	ITEST	LASTEP			

4 (8E10.4)

ZUI	ZUE	ZULAST	FRA	ZEND	ZOUT	PRESS	POWER

5 (8 (12, 8X))

NBP	ICURV	NRCVI	NRCVE				

6 (8E10.4)

RI ₁	XI ₁	RI ₂	XI ₂	RI ₃	XI ₃	RI ₄	XI ₄ ...

(8E10.4)

RE ₁	XE ₁	RE ₂	XE ₂	RE ₃	XE ₃	RE ₄	XE ₄ ...

7 (8E10.4)

ZI ₁	RCVI ₁	ZI ₂	RCVI ₂	ZI ₃	RCVI ₃	ZI ₄	RCVI ₄ ...

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(8E10.4)

ZE₁ RCVE₁ ZE₂ RCVE₂ ZE₃ RCVE₃ ZE₄ RCVE₄...

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RA RB RD (8E10.4)

8

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NAMELIST

9

\$INPUT							\$
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UA UD (8E10.4)

10

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F2A F2D TA TD T_{wall} (8E10.4)

11

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NIN (I2)

12

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Y VALUES (8E10.4)

13

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U VALUES (8E10.4)

14

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15

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TEMPERATURE VALUES (8E10.4)

16

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17

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18

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19

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20

TURBULENT KINETIC ENERGY						(8E10.4)	

21

TURBULENCE LENGTH SCALE						(8E10.4)	

TRANSITION MIXING MODEL
INPUT SHEET DESCRIPTION

Card Set	Description
1	Case title cards
2	N - Number of cross stream intervals KRAD - 00 For plane geometry 01 For axisymmetric geometry MASSTR - 00 Always ISWRL - 00 Always MODEL - 01 Laminar viscosity 02 Two-equation K-E viscosity model INERT - 01 Always IFLUX - 00 Walls are adiabatic 01 Wall temperature is specified ITEMP - 00 Isothermal 01 Enthalpy solved
3	TAT - Number of steps between printout of output variables NPROF - Number of steps between printout of output variables and profiles NPLOT - Number of steps between line-printer plots ITEST - 00 No extra printout 01 Extra printout LASTEP - Maximum number of marching steps
4	ZUI - Initial Z-location on inner boundary ZUE - Initial Z-location on outer boundary ZULAST - Final Z location FRA - Fraction of grid height to be used as step size ZEND - Z-location of end of inner wall ZOUT - Z-location of end of outer wall PRESS - Pressure POWER - Control for node spacing
5	NBP - Number of boundary pairs ICURV - 00 No radius of curvature effects 01 Radius of curvature effects NRCVI - Number of radius of curvature points specified on inner boundary NRCVE - Number of radius of curvature points specified on outer boundary
6	RI ₁ - Inner boundary radius XI ₁ - Inner boundary X-location RE ₁ - Outer boundary radius XE ₁ - Outer boundary X-location

- 7 ZI_1 - Z-location on inner boundary at which
 radius of curvature is specified
 $RCVI_1$ - Radius of curvature of inner boundary
 ZE_1 - Z-location on outer boundary at which
 radius of curvature is specified
 $RCVE_1$ - Radius of curvature of outer boundary
- 8 RA - Radius of axis of symmetry
 RB - Inner boundary radius at initial Z-location
 RD - Outer boundary radius at initial Z-location
- 9 Namelist, see page 164 for variables
- 10 UA - Axial velocity at "free" inner boundary
 UD - Axial velocity at "free" outer boundary
- 11 F2A - Fuel mass fraction at "free" inner boundary
 F2D - Fuel mass fraction at "free" outer boundary
 TA - Temperature at "free" inner boundary
 TD - Temperature at "free" outer boundary
 TWALL - Wall temperature
- 12 NIN - Number of points on input initial profile
- 13 Y values of input initial profile (NIN values)
- 14 U values of input initial profile
- 15 Blank card(s)
- 16 Temperature values of input initial profile
- 17 Blank card(s)
- 18 Blank card(s)
- 19 Blank card(s)
- 20 RE values of input initial profile
 (Read only if KREAD = 1)
- 21 μ values of input initial profile
 (Read only if KREAD = 1)

TRANSITION MIXING MODEL
NAMELIST INPUT

VBL	Value	Description
IUTRAP	2	Test for negative U's, see STRIDE(2)
ULIM	0.05	Entrainment control
PEILIM	0.01	Max. fractional mass flow change per step
AFAC	0.2	Relaxation on duct area deviation
AEXDLM	0.02	Max. duct area fractional deviation per step
NOVEL	2	01 - U not solved for, 02 - solve for U
ARCON1	28500	Activation energy divided by gas constant for fuel reaction
PREXP1	5E + 15	Preexponent for fuel reaction
CR1	6.0	Eddy break-up constant for fuel reaction
ARCON2	12500	Activation energy divided by gas constant for CO reaction
PREXP2	6E + 8	Preexponent for CO reaction
CR2	6.0	Eddy break-up constant for CO reaction
CP	1255	Specific heat
IDPDX	01	01 - Std genmix pressure grad. calculation 02 - "grid filling duct" version
C1	1.42	Turb. Constant
C2	1.92	Turb. Constant
CD	0.09	Turb. Constant
C2VT	0.36	Turb. Constant
AKFAC	0.03	Factor for internally generated turb. kinetic energy profiles, $KE = AKFAC * U^2$
ALFAC	0.02	Factor for internally generated length scale profiles, $l_m = ALFAC * \Delta Y$
PREF	--	Turbulent Prandtl numbers
CX	1.0	Molecular carbon value of fuel
HY	4.0	Molecular hydrogen value of fuel
HFU	4E + 7	Heat of combustion of fuel
MODER	2	01 - Kinetic only, 02 - Kinetic + Diffusion
ITHIN	1	Thins output profiles by printing every ITHIN value
KREAD	0	00 - KE & l_m profiles generated internally 01 - KE & l_m read in

EMISSIONS MODEL INPUT SHEET

CASE TITLE CARD (12A6)

1

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CASE TITLE CARD (12A6)

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(8 (I2, 8X))

N KRAD MASSTR ISWRL MODEL INERT IFLUX ITEMP

2

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NSTAT NPROF NPLOT ITEST LASTEP (5 (I5, 5X))

3

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XU XULAST FRA XEND XOUT PRESS POWER (8E10.4)

4

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NBP (I2)

5

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X₁ RI₁ RE₁ X₂ RI₂ RE₂ X₃ RI₃

6

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RE₃ X₄ RI₄ RE₄ X₅ RI₅ RE ... (8E10.4)

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RA RB RD (8E10.4)

7

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NAMELIST

8	\$INPUT							\$
9	UA	UD	VTA	VTD				(8E10.4)
10	F2A	F2D	TA	TD	T _{wall}			(8E10.4)
11	NIN	NUI	NUE	NVI	NVE			(5 (12, 8X))
12	Y VALUES							(8E10.4)
13	U VALUES							(8E10.4)
14	V _θ VALUES							(8E10.4)
15	TEMPERATURE VALUES							(8E10.4)
16	MFU VALUES							(8E10.4)
17	MCO ₂ VALUES							(8E10.4)

18 MCO VALUES (8E10.4)

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19 MOX VALUES (8E10.4)

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20 MH_2O VALUES (8E10.4)

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21 MH_2 VALUES (8E10.4)

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22 TURBULENT KINETIC ENERGY (8E10.4)

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23 TURBULENCE LENGTH SCALE (8E10.4)

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———— SKIP FOLLOWING CARD SET IF NUI = 0 ————

24 X - LOC. OF INTERNAL COOLING SLOTS (8E10.4)

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25 LIP LENGTH OF INTERNAL COOLING SLOTS (8E10.4)

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26 U - VELOCITY OF INTERNAL COOLING SLOTS (8E10.4)

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27 V_T - VELOCITY OF INTERNAL COOLING SLOTS (8E10.4)

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TEMPERATURE OF INTERNAL COOLING SLOTS (8E10.4)

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FLOW RATE OF INTERNAL COOLING SLOTS (8E10.4)

29

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SLOT HEIGHT OF INTERNAL COOLING SLOTS (8E10.4)

30

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SLOT TO METERING AREA RATIO FOR INT SLOTS (8E10.4)

31

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SKIP FOLLOWING CARD SET IF NUE = 0

X-LOC OF EXTERNAL COOLING SLOTS (8E10.4)

32

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LIP LENGTH OF EXTERNAL COOLING SLOTS (8E10.4)

33

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U - VELOCITY OF EXTERNAL COOLING SLOTS (8E10.4)

34

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V_T - VELOCITY OF EXTERNAL COOLING SLOTS (8E10.4)

35

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36 TEMPERATURE OF EXTERNAL COOLING SLOTS (8E10.4)

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37 FLOW RATE OF EXTERNAL COOLING SLOTS (8E10.4)

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38 SLOT HEIGHT OF EXTERNAL COOLING SLOTS (8E10.4)

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39 SLOT TO METERING AREA RATIO FOR EXT SLOTS (8E10.4)

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SKIP FOLLOWING CARD SET IF NVI = 0

40 X - LOC OF INTERNAL RADIAL INJECTION (8E10.4)

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41 U - VELOCITY OF INTERNAL RADIAL INJECTION (8E10.4)

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42 V - VELOCITY OF INTERNAL RADIAL INJECTION (8E10.4)

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43 VT - VELOCITY OF INTERNAL RADIAL INJECTION (8E10.4)

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TEMPERATURE OF INTERNAL RADIAL INJECTION (8E10.4)

44

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FLOW RATE OF INTERNAL RADIAL INJECTION (8E10.4)

45

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SKIP FOLLOWING CARD SET IF NVE = 0

X - LOC. OF EXTERNAL RADIAL INJECTION (8E10.4)

46

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U - VELOCITY OF EXTERNAL RADIAL INJECTION (8E10.4)

47

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V - VELOCITY OF EXTERNAL RADIAL INJECTION (8E10.4)

48

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VT - VELOCITY OF EXTERNAL RADIAL INJECTION (8E10.4)

49

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TEMPERATURE OF EXTERNAL RADIAL INJECTION (8E10.4)

50

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FLOW RATE OF EXTERNAL RADIAL INJECTION (8E10.4)

51

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EMISSIONS MODEL
INPUT SHEET DESCRIPTION

Card Set	Description
1	Case title cards
2	N - Number of cross stream intervals KRAD - 00 For plane geometry 01 For axisymmetric geometry MASSTR - 00 Species equations not solved 01 Species equations solved ISWRL - 00 Swirl velocity not solved 01 Swirl velocity solved MODEL - 01 Laminar viscosity 02 Two-equation, K-E, viscosity model INERT - 01 Species are inert 02 Species will react IFLUX - 00 Wall adiabatic 01 Wall temperature specified ITEMP - 00 Isothermal 01 Enthalpy solved
3	NSTAT - Number of steps between printout of output variables NPROF - Number of steps between printout of output variables and profiles NPLOT - Number of steps between line printer plots ITEST - 00 No extra printout 01 Extra printout LASTEP - Maximum number of marching steps
4	XU - Initial X location XULAST - Final X location FRA - Fraction of grid height to be used as step size XEND - X location of end of inner wall XOUT - X location of end of outer wall PRESS - Pressure POWER - Control for node spacing
5	NBP - Number of boundary pairs
6	X _i - X location at which boundary is specified RI _i - Inner boundary radius RE _i - Outer boundary radius
7	RA - Radius of axis of symmetry RB - Inner boundary radius at initial X-location RD - Outer boundary radius at initial X-location

- 8 Namelist, see page 174 for variables
- 9 UA - Axial velocity at "free" inner boundary
 UD - Axial velocity at "free" outer boundary
 VTA - Tang. velocity at "free" inner boundary
 VTD - Tang. velocity at "free" outer boundary
- 10 F2A - Fuel mass fraction at "free" inner boundary
 F2D - Fuel mass fraction at "free" outer boundary
 TA - Temperature at "free" inner boundary
 TD - Temperature at "free" outer boundary
 TWALL - Wall temperature
- 11 NIN - Number of points on the input initial profile
 NUI - Number of cooling slots on inner boundary
 NUE - Number of cooling slots on outer boundary
 NVI - Number of radial injection points on inner boundary
 NVE - Number of radial injection points on outer boundary
- 12 Y values of input initial profile (NIN values)
- 13 U values of input initial profile
- 14 V_{θ} values of input initial profile
- 15 Temperature values of input initial profile
- 16 M_{F_2} values of input initial profile
- 17 M_{CO_2} values of input initial profile
- 18 M_{CO} values of input initial profile
- 19 M_{O_2} values of input initial profile
- 20 M_{H_2O} values of input initial profile
- 21 M_{H_2} values of input initial profile
- 22 KE values of input initial profile
 (Read only if KREAD = 1)
- 23 μ values of input initial profile
 (Read only if KREAD = 1)
- 24 to 31 Information pertaining to cooling slots on inner boundary

- 32 to 39 Information pertaining to cooling slots. on outer boundary
- 40 to 45 Information pertaining to radial injections on inner boundary
- 46 to 51 Information pertaining to radial injections on outer boundary

EMISSIONS MODEL
NAMELIST INPUT

VBL	Value	Description
IUTRAP	2	Test for negative U's, see STRIDE(2)
ULIM	0.05	Entrainment control
PEILIM	0.01	Max. fractional mass flow change per step
AFAC	0.2	Relaxation on duct area deviation
AEXDLM	0.02	Max. duct area fractional deviation per step
NOVEL	2	01 - U not solved for, 02 - Solve for U
ARCON1	28500	Activation energy for fuel reaction
PREXP1	5E + 15	Preexponent for fuel reaction
CR1	6.0	Eddy breakup constant for fuel reaction
CP	1048	Specific heat
IPDDX	1	01 - Std genmix pressure grad. calculation 02 - "grid filling duct" version
C1	1.42	Turb. Constant
C2	1.92	Turb. Constant
CD	0.09	Turb. Constant
C2VT	0.36	Turb. Constant
AKFAC	0.03	Factor for internally generated kinetic energy profiles, $KE = AKFAC * U$
ALFAC	0.02	Factor for internally generated length scale profiles, $l_m = ALFAC * \Delta Y$
PREF	--	Turbulent Prandtl numbers
CX	1.0	Molecular carbon value of fuel
HY	4.0	Molecular hydrogen value of fuel
HFU	-49317	Heat of formation of fuel
MODER	2	1 - Kinetic only, 02 - Kinetic + Diffusion
ITHIN	1	Thins output profiles by printing every ITHIN value
KREAD	0	0 - K & E profiles generated internally 1 - K & l_m read in
URAT	0.83	Cooling slot velocity shape factor
TERM1	0.1	Control on specie equation step size
TERM2	1.E-4	Control on specie equation step size
ISMAX	500	Maximum specie equation steps
EFU	1.0	Power on fuel in fuel reaction rate
E _{O2}	0.5	Power on O ₂ in fuel reaction rate
E _{H2O}	0.5	Power on H ₂ O in fuel reaction rate
ERO	2.0	Power on density in fuel reaction rate
EPR	0.0	Power on pressure in fuel reaction rate

FUEL INSERTION MODEL INPUT

1 TITLE 80

	* **	PRIM FLOW	SEC. FLOW	CONE ANGLE	AIR SHROUD EFF AREA	PRIM ORIFICE DIA, IN	SEC ORIFICE DIA, IN		
		NO. (JP4)	NO. (JP4)	DEG	IN ²	IN	IN		
		PPH/ $\sqrt{\text{PSI}}$							
1	→ 6 → 11	21	31	41	51	61	71	80	
2									

*ATOMIZER TYPE: 00001 = SIMPLEX
 00002 = DUAL ORF
 00003 = AIR BLAST

**AIR ASSIST: 00001 = NO ASSIST
 00002 = WITH ASSIST

*FOR ATOM TYPE = 00002 (DUAL ORIF) ONLY (LEAVE OUT FOR OTHERS)
 INPUT SECONDARY FLOW SCHEDULE

W_s = SECONDARY FUEL FLOW, LB/HR
 ΔP_s = $\Delta P_{\text{SEC. ORIF}}$ + $\Delta P_{\text{FLOW DIVIDER VALVE}}$ P/D

CRACK POINT FLOW

	1 W_{s1}	11 W_{s2}	21 W_{s3}	31 W_{s4}	41 W_{s5}	51
3						

CRACK PRESSURE

	1 ΔP_1	11 ΔP_2	21 ΔP_3	31 ΔP_4	41 ΔP_5	51
4						

*FOR ATOM TYPE = 00003 (AIRBLAST) ONLY (LEAVE OUT FOR OTHERS)

	FILMING NOZZLE DIA. IN.	AIRFLOW LB/SEC	FLOW AREA IN ²	AIR TEMP °R
1	11	21	31	
5				

 *** AIR FUEL FUEL AIR AIR
 FUEL FLOW TEMP FLOW FUEL ASSIST ASSIST
 TYPE OPTION °R LB/HR PSI ΔP ΔP TEMP
 1 → 6 → 11 21 31 41 51 61 71

6

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FUEL 0.00002=JP5 *AIR }
 TYPE 0.00004=JP4 FLOW } 0.00001=UNIFORM GAS STREAM
 OPTION } 0.00002=2-D FIELD OPTION

T_{GAS}, °R V_{GAS}, P_{GAS}, X_{MAX}, Y_{MAX}, ← UNIFORM STREAM
 1 11 FPS 21 PSIA 31 IN. 4 °R 51 OPTION

7

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↑ ↑ ↑ ↑ ↑ ← Z-D FIELD OPTION
 X_{NOZ} ε Y_{NOZ} ε P_{GAS} X_{MAX} Y_{MAX}
 IN. IN. PSIA IN. IN.

CARDS 8 THROUGH 13 SKIPPED IF AIR FLOW OPTION = 00001

IN = NO. OF X-DIR POINTS IN 2-D FIELD
 JN = NO. OF Y-DIR POINTS
 IN JN (2(I2, 8X))

8

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X VALUES (8E10.4) X - LOCATIONS OF GRID POINTS (FT)

9

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Y VALUES (8E10.4) Y - LOCATIONS OF GRID POINTS (FT)

10

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READ ALONG + X LINES STARTING WITH
 U VALUES (8E10.4) SMALLEST Y VALUE (FT/SEC)

11

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READ ALONG + X LINES STARTING WITH
 V VALUES (SE10.4) SMALLEST Y VALUE (FT/SEC)

12

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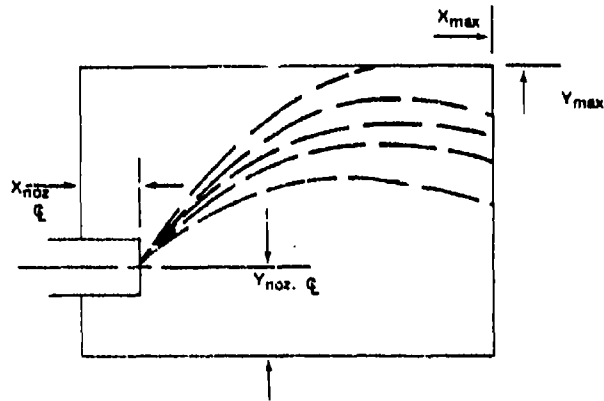
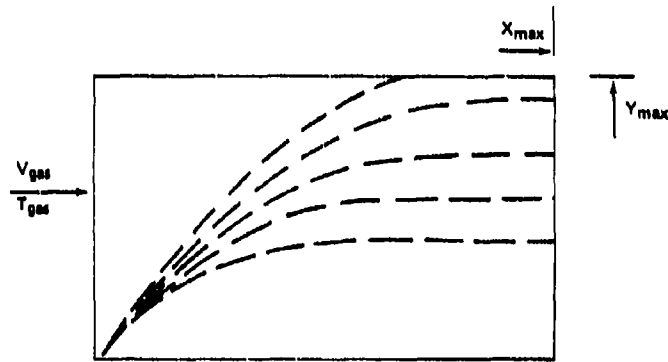
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READ ALONG + X LINES STARTING WITH
 T VALUES (SE10.4) SMALLEST Y VALUE (DEG. R)

13

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PDCPAR MAIN 73/74 OPT=0 TRACE PAGE 2
 SFM 4-0-6439 08/24/78 02.58.39
 80 CALL PRODUCE(1)
 IF (IPIC-EG) CALL PICT01
 CALL WAP(14) GO TO 70
 CALL WAP(14)
 CALL WAP(14)
 IF (IPIC-EG) CALL PICT01
 CALL WAP(14)
 CALL WAP(14)
 GO TO 70
 C
 80 WRITE (UNIT,MESSAGES
 75 FORMAT ('*****
 1 *****')
 (END ,ITERZ ,15)
 81
 82
 83
 84
 85
 86
 87
 88
 89
 90
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 92
 93
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 95
 96
 97
 98
 99

1	SUBROUTINE TAPEINT)	TA	2
	COMMON/CI CYL,IND,YMO,NEW,VOL,VOL,TOV,TOV,TOB(000),YOB(000),XG,YG,	CBI	3
	1 SIZE,XI,XTL,MOVE,VMW,DATA(12),MVL(12)	CBI	4
5	COMMON/IRAPUCZ,P,DLFPA,IG,APIC,TAN,COB,ME,LA,TITLE(0),RELTUT,	CBI	5
	1 TITLE,COB,NEW,LI,OR,DR,PI,OD,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,	CBI	6
	1 ISL,TOB(0),E,DRUG,MF,COB,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,	CBI	7
	1 COMMON/OUTPUT/PTIS,CAN,COB,L,CAN,COM,SI,OR,SI,OR,SI,OR,SI,OR,SI,	CBI	8
10	1 AT(000) BLANK(000) BLK(000) EX(000) EX(000) EX(000) EX(000) EX(000),	CBI	9
	1 J(000) AT(000),PT(000),PT(000),PT(000),PT(000),PT(000),PT(000),	CBI	10
	1 C(000) A(000) C(000) A(000) C(000) A(000) C(000) A(000) C(000),	CBI	11
15	1 V(ET(000) AT(000) RE(000) CUM(000) CUM(000) CUM(000) CUM(000),	CBI	12
	1 COMMON/CODINT/IM,ROUT,RE,AT,IVE,SUFF,SUFF	TA	13
	1 EMBY,AT	TA	14
	1 RE(000) IM,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,VE,	TA	15
	1 READ(INT) OP,OLE,SK,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,	TA	16
20	1 RETYP,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,	TA	17
	1 IDBUG,MP,PI,C,ABEY,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,	TA	18
	1 HEAD(INT) PDIS,COB,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,	TA	19
25	1 ED,ME,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,	TA	20
	1 HEAD(INT) TTER,SUFF,SUFF	TA	21
	1 RETURN	TA	22
	1 END	TA	23
30	EMPTY TAPE	TA	24
	1 WRITE(INT) END,END,END,END,END,END,END,END,END,END,END,END,	TA	25
	1 WRITE(INT) MOVE,DATA,DATA,DATA,DATA,DATA,DATA,DATA,DATA,DATA,	TA	26
	1 WRITE(INT) OP,OLE,SK,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,SC,	TA	27
35	1 RETYP,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,CO,	TA	28
	1 IDBUG,MP,PI,C,ABEY,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,SIG,	TA	29
	1 HEAD(INT) PDIS,COB,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,OR,SI,	TA	30
	1 ED,ME,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,IM,	TA	31
	1 HEAD(INT) TTER,SUFF,SUFF	TA	32
	1 RETURN	TA	33
40	1 END	TA	34


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00 IF (CASE.EQ.1)MELMELI1=MELM
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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM

140 I Z0C0 140 FIELD 0874 OF A COMPRESSION DESCRIPTION SHOULD BE AS LARGE AS THE WINDOW SPECIFIED FOR THAT DESCRIPTION.

SUBROUTINE INLET TS/TA OPT=0 TRACE

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1  CHARACTER*(*) NAME, COM, MEI, TITLE, IOT, MEI, TOT,
2  TITLE, COM, MEI, IOT, MEI, TOT, MEI, TOT, MEI, TOT,
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1 SUBROUTINE MEX(L,M,N)
2   DIMENSION M(100),N(100)
3   M=0
4   N=0
5   DO 10 I=1,M
6     DO 10 J=1,N
7       M(I,J)=0
8       N(I,J)=0
9     END DO
10  END DO
11  M=1
12  N=1
13  M=2
14  N=2
15  M=3
16  N=3
17  M=4
18  N=4
19  M=5
20  N=5
21  M=6
22  N=6
23  M=7
24  N=7
25  M=8
26  N=8
27  M=9
28  N=9
29  M=10
30  N=10
31  END

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1 SUBROUTINE JET(L,L1,L2,L3)
2 COMMON/CPD/PT,MT,MTOUT,IPRAT,ITR,SUCCESS
3 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
4 IF (L1.LT.1) GO TO 10
5 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
6 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
7 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
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60 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
61 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
62 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
63 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
64 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
65 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
66 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
67 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
68 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
69 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
70 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
71 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
72 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
73 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
74 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)
75 CALL MPOB(PT,MT,MTOUT,IPRAT,ITR,SUCCESS)

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1 SUBROUTINE DCOEF (M,N,JH,BLI,MMOTYP,CDJ,AJET)
2   W=JH*MMOTYP
3   IF (LTCOS(1)-.4) GO TO 10
4   IF (LTCOS(1)-.4) GO TO 10
5   IF (LTCOS(1)-.4) GO TO 10
6   IF (LTCOS(1)-.4) GO TO 10
7   IF (LTCOS(1)-.4) GO TO 10
8   IF (LTCOS(1)-.4) GO TO 10
9   IF (LTCOS(1)-.4) GO TO 10
10  PI=3.141592654
11  RAD=51.297795131
12  IF (M-CDJ) GO TO 70
13  IF (M-CDJ) GO TO 70
14  BLD=ACOS(1)-M/2.100AD
15  B=ACOS(1)-M/2.11
16  C=1.0-M/2.
17  C=1.0-M/2.
18  C=1.0-M/2.
19  C=1.0-M/2.
20  C=1.0-M/2.
21  C=1.0-M/2.
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63  C=1.0-M/2.
64  C=1.0-M/2.
65  C=1.0-M/2.
66  C=1.0-M/2.
67  C=1.0-M/2.
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69  C=1.0-M/2.
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79  C=1.0-M/2.
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81  C=1.0-M/2.
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87  C=1.0-M/2.
88  C=1.0-M/2.
89  C=1.0-M/2.
90  C=1.0-M/2.
91  C=1.0-M/2.
92  C=1.0-M/2.
93  C=1.0-M/2.
94  C=1.0-M/2.
95  C=1.0-M/2.
96  C=1.0-M/2.
97  C=1.0-M/2.
98  C=1.0-M/2.
99  C=1.0-M/2.
100 C=1.0-M/2.

```

SUBROUTINE BELTO 7374 OPT=0 TRACE STN 4-6-439 00724778 29.37.39 PAGE 1
 1 2
 5 3
 10 4
 15 5
 20 6
 30 7
 40 8
 50 9
 END BELTO

1 SUBROUTINE BELTO(B,A,M,N,C,L,M,T,X,D,T,Y,P,R,I,C,E,F,F,A,R,M,E)1
 2 C=0
 3
 4 B=1.0
 5
 6
 7
 8
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 11
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 49
 50
 END BELTO
 C

```

SUBROUTINE ARM
  73/74  OPT=3 TRACE          STM 4.00030          09/24/79  00.55.39          PAGE 1
1  SUBROUTINE ARM(DAA,DTT,DF,DD,YS,Se,PI,Emp,EMZ)
  2  PA=2
  3  DIMENSION(SM-1,1/2,0EMB/1,Emp)
  4  PI=EMZ
  5  PI=EMZ
  6  PI=EMZ
  7  PI=EMZ
  8  PI=EMZ
  9  PI=EMZ
 10  PI=EMZ
 11  PI=EMZ
 12  PI=EMZ
 13  PI=EMZ
 14  PI=EMZ
 15  C
 16  C
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 96  C
 97  C
 98  C
 99  C
100 C

```


SUBROUTINE NAME	7374	CPI-05 TRACE	STM 4.0-430	08/26/78	09-58-30	PAGE	1
1		<pre> SUBROUTINE KAREAINC-01.07.01.09-0.05.01 K=3145605 P=00110-11/2-1 AS=2.01000100110100-0110100-0111 RETURN END KAREIN </pre>					
5				<pre> NA NA NA NA NA NA NA </pre>			<pre> 1 1 1 1 1 1 10 </pre>


```

1 SUBROUTINE PRODU(P)
2 CUMMAREA=0.0
3 L=0.0
4 L=0.0
5 L=0.0
6 L=0.0
7 L=0.0
8 L=0.0
9 L=0.0
10 L=0.0
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14 L=0.0
15 L=0.0
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38 L=0.0
39 L=0.0
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41 L=0.0
42 L=0.0
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56 L=0.0
57 L=0.0
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59 L=0.0
60 L=0.0
61 L=0.0
62 L=0.0
63 L=0.0
64 L=0.0
65 L=0.0

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1 SUBROUTINE PICTURE
5 1 SIZE=INT (PICTURE)
5 2 COMBIN=INT (PICTURE)
5 3 COMB=INT (PICTURE)
10 1 ARROW=INT (PICTURE)
10 2 WID=INT (PICTURE)
10 3 HGT=INT (PICTURE)
15 1 COMB=INT (PICTURE)
15 2 WID=INT (PICTURE)
15 3 HGT=INT (PICTURE)
20 1 COMB=INT (PICTURE)
20 2 WID=INT (PICTURE)
20 3 HGT=INT (PICTURE)
25 1 COMB=INT (PICTURE)
25 2 WID=INT (PICTURE)
25 3 HGT=INT (PICTURE)
30 1 COMB=INT (PICTURE)
30 2 WID=INT (PICTURE)
30 3 HGT=INT (PICTURE)
35 1 COMB=INT (PICTURE)
35 2 WID=INT (PICTURE)
35 3 HGT=INT (PICTURE)
40 1 COMB=INT (PICTURE)
40 2 WID=INT (PICTURE)
40 3 HGT=INT (PICTURE)
45 1 COMB=INT (PICTURE)
45 2 WID=INT (PICTURE)
45 3 HGT=INT (PICTURE)
50 1 COMB=INT (PICTURE)
50 2 WID=INT (PICTURE)
50 3 HGT=INT (PICTURE)
55 1 COMB=INT (PICTURE)
55 2 WID=INT (PICTURE)
55 3 HGT=INT (PICTURE)
60 1 COMB=INT (PICTURE)
60 2 WID=INT (PICTURE)
60 3 HGT=INT (PICTURE)
65 1 COMB=INT (PICTURE)
65 2 WID=INT (PICTURE)
65 3 HGT=INT (PICTURE)
70 1 COMB=INT (PICTURE)
70 2 WID=INT (PICTURE)
70 3 HGT=INT (PICTURE)
75 1 COMB=INT (PICTURE)
75 2 WID=INT (PICTURE)
75 3 HGT=INT (PICTURE)

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80 15 (L)E TRV(111) 58-1(46) 198111100000
81 16 (L)E TRV(111) 58-1(46) 198111100000
82 17 (L)E TRV(111) 58-1(46) 198111100000
83 18 (L)E TRV(111) 58-1(46) 198111100000
84 19 (L)E TRV(111) 58-1(46) 198111100000
85 20 (L)E TRV(111) 58-1(46) 198111100000
86 21 (L)E TRV(111) 58-1(46) 198111100000
87 22 (L)E TRV(111) 58-1(46) 198111100000
88 23 (L)E TRV(111) 58-1(46) 198111100000
89 24 (L)E TRV(111) 58-1(46) 198111100000
90 25 (L)E TRV(111) 58-1(46) 198111100000
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96 31 (L)E TRV(111) 58-1(46) 198111100000
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98 33 (L)E TRV(111) 58-1(46) 198111100000
99 34 (L)E TRV(111) 58-1(46) 198111100000
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103 38 (L)E TRV(111) 58-1(46) 198111100000
104 39 (L)E TRV(111) 58-1(46) 198111100000
105 40 (L)E TRV(111) 58-1(46) 198111100000
106 41 (L)E TRV(111) 58-1(46) 198111100000
107 42 (L)E TRV(111) 58-1(46) 198111100000
108 43 (L)E TRV(111) 58-1(46) 198111100000
109 44 (L)E TRV(111) 58-1(46) 198111100000
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128 63 (L)E TRV(111) 58-1(46) 198111100000
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130 65 (L)E TRV(111) 58-1(46) 198111100000
131 66 (L)E TRV(111) 58-1(46) 198111100000
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136 71 (L)E TRV(111) 58-1(46) 198111100000
137 72 (L)E TRV(111) 58-1(46) 198111100000
138 73 (L)E TRV(111) 58-1(46) 198111100000
139 74 (L)E TRV(111) 58-1(46) 198111100000
140 75 (L)E TRV(111) 58-1(46) 198111100000
141 76 (L)E TRV(111) 58-1(46) 198111100000
142 77 (L)E TRV(111) 58-1(46) 198111100000
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144 79 (L)E TRV(111) 58-1(46) 198111100000
145 80 (L)E TRV(111) 58-1(46) 198111100000
146 81 (L)E TRV(111) 58-1(46) 198111100000
147 82 (L)E TRV(111) 58-1(46) 198111100000
148 83 (L)E TRV(111) 58-1(46) 198111100000
149 84 (L)E TRV(111) 58-1(46) 198111100000
150 85 (L)E TRV(111) 58-1(46) 198111100000
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SUBROUTINE PICTURE 73/76 OPT=0 TRACE

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155      X=K(11)-SPAC2
      CALL NUMBER (H,Y,O,I,DM,0,0,21)
      C 1065 CONTINUE      RESTORE VALUE OF R'S AND X'S
      C
      DO 1070 I=1,NEL
      X(11)=K(11)+RNEAN
      X(12)=K(12)+RNEAN
      X(13)=K(13)+RNEAN
      X(14)=K(14)+RNEAN
      X(15)=K(15)+RNEAN
      X(16)=K(16)+RNEAN
      X(17)=K(17)+RNEAN
      X(18)=K(18)+RNEAN
      C 1070 CONTINUE      DRAW ROSES
      C
      CALL BODES
      RETURN
      END COMDT
170      C

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SUBROUTINE BOXES
1  SUBROUTINE BOXES
2  DIMENSION BOXES(100), BOXES2(100), BOXES3(100), BOXES4(100), BOXES5(100),
3  BOXES6(100), BOXES7(100), BOXES8(100), BOXES9(100), BOXES10(100),
4  BOXES11(100), BOXES12(100), BOXES13(100), BOXES14(100), BOXES15(100),
5  BOXES16(100), BOXES17(100), BOXES18(100), BOXES19(100), BOXES20(100),
6  BOXES21(100), BOXES22(100), BOXES23(100), BOXES24(100), BOXES25(100),
7  BOXES26(100), BOXES27(100), BOXES28(100), BOXES29(100), BOXES30(100),
8  BOXES31(100), BOXES32(100), BOXES33(100), BOXES34(100), BOXES35(100),
9  BOXES36(100), BOXES37(100), BOXES38(100), BOXES39(100), BOXES40(100),
10 BOXES41(100), BOXES42(100), BOXES43(100), BOXES44(100), BOXES45(100),
11 BOXES46(100), BOXES47(100), BOXES48(100), BOXES49(100), BOXES50(100),
12 BOXES51(100), BOXES52(100), BOXES53(100), BOXES54(100), BOXES55(100),
13 BOXES56(100), BOXES57(100), BOXES58(100), BOXES59(100), BOXES60(100),
14 BOXES61(100), BOXES62(100), BOXES63(100), BOXES64(100), BOXES65(100),
15 BOXES66(100), BOXES67(100), BOXES68(100), BOXES69(100), BOXES70(100),
16 BOXES71(100), BOXES72(100), BOXES73(100), BOXES74(100), BOXES75(100),
17 BOXES76(100), BOXES77(100), BOXES78(100), BOXES79(100), BOXES80(100),
18 BOXES81(100), BOXES82(100), BOXES83(100), BOXES84(100), BOXES85(100),
19 BOXES86(100), BOXES87(100), BOXES88(100), BOXES89(100), BOXES90(100),
20 BOXES91(100), BOXES92(100), BOXES93(100), BOXES94(100), BOXES95(100),
21 BOXES96(100), BOXES97(100), BOXES98(100), BOXES99(100), BOXES100(100)
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08 CALL SYMBOLS,V..I..24MISICE ELEMENT IS 8 - BELFED FLOW EXTRACTED.O  
09  
10 CALL SYMBOLS,V..I..24MISICE ELEMENT IS 9 - FIRED HOLE DIAMETER.O  
11  
12 CALL SYMBOLS,V..I..24MISICE ELEMENT IS 10 - FIRED FLOW PERCENT .O  
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08/24/78 *CBC CV174 SCOPE VER3.3.4 DATED 09/16/76
 09.50.31.0147TIP
 09.50.31.SEQUENCE=0147T
 09.50.31.200-CMS5806-1119.
 09.50.31-USER=REPLD5833DEPT-93-352-EMD-3409-2458
 09.50.31-30-05-0501-PRDGRAB-0.
 09.50.32-BARME9, DAYFILE, 01M 27
 09.50.32. XXXI XXX X F
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 09.50.32.
 09.50.32-REQUEST, DLO, PE, WRN=639276, LR=AVDATA, ID=0
 09.50.32-ETNOLDS, FILES=48-67.
 09.50.32-PENIND(OL0)
 09.50.34-UPDATE(OL0, OL044)
 09.50.38-UPDATE COMPLETE.
 09.50.38-CUP(OL0, PD=0)
 09.50.10-12,499 CP SECONDS COMPILATION TIME
 09.50.10-TOTAL SRU 63,723
 09.50.10-TOTAL CP 11,647 SEC.
 09.50.10-TOTAL PP 30,770 SEC.
 09.50.10-TOTAL IO 11,615 SEC.
 09.50.10-TOTAL RA+I CALLS 566.

0147TIP //// END OF LIST //// EST 33
 0147TIP //// END OF LIST //// EST 33


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200 I(I)OCLEMO(I)
201 DO 205 I=1,N
205 I(I)OCLEMO(I)
206 DO 220 I=1,M
215 I(I)OCLEMO(I),IPLAR
220 CONTINUE
221 I(I)OCLEMO(I)
C -----
20 27 J=JMP1
21 28 J=JMP2
22 29 J=JMP3
23 30 J=JMP4
24 31 J=JMP5
JMP1=1
JMP2=1
JMP3=1
JMP4=1
JMP5=1
25 32 I=1,100
26 33 I=1,100
27 34 I=1,100
28 35 I=1,100
29 36 I=1,100
30 37 I=1,100
31 38 I=1,100
32 39 I=1,100
33 40 I=1,100
34 41 I=1,100
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103 110 I=1,100
104 111 I=1,100
105 112 I=1,100
106 113 I=1,100
107 114 I=1,100
108 115 I=1,100

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155 I=I+1;CLAS(TEMP(U))
160 I=I+1;CLAS(TEMP(U))
165 I=I+1;CLAS(TEMP(U))
170 I=I+1;CLAS(TEMP(U))
175 I=I+1;CLAS(TEMP(U))
180 I=I+1;CLAS(TEMP(U))
185 I=I+1;CLAS(TEMP(U))
190 I=I+1;CLAS(TEMP(U))
195 I=I+1;CLAS(TEMP(U))
200 I=I+1;CLAS(TEMP(U))
205 I=I+1;CLAS(TEMP(U))
210 I=I+1;CLAS(TEMP(U))
215 I=I+1;CLAS(TEMP(U))
220 I=I+1;CLAS(TEMP(U))
225 I=I+1;CLAS(TEMP(U))
230 I=I+1;CLAS(TEMP(U))

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233 CHAPTER 8
      IF (STEP-AS) STEP, 400000
      DO WRITE (6,13) (STEP,33)
      C -----
      STOP
      END
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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM

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CONTROL VARIABLE IN COMMON OR EQUIVALENT. OPTIMIZATION MAY BE OMITTED.
 AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
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 CONTROL VARIABLE IN COMMON OR EQUIVALENT. OPTIMIZATION MAY BE OMITTED.

DR/26/78 14-08-66

FM 4.6-439

SUBROUTINE INITIAL 73774 OPT-0 TRACE

1	SUBROUTINE INITIAL COMMON R1(10) R2(10) R3(10) R4(10) R5(10) R6(10) R7(10) R8(10) R9(10) R10(10)			AL	CONF	1
5	COMMON R11(10) R12(10) R13(10) R14(10) R15(10) R16(10) R17(10) R18(10) R19(10) R20(10)			AL	CONF	5
10	COMMON R21(10) R22(10) R23(10) R24(10) R25(10) R26(10) R27(10) R28(10) R29(10) R30(10)			AL	CONF	10
15	COMMON R31(10) R32(10) R33(10) R34(10) R35(10) R36(10) R37(10) R38(10) R39(10) R40(10)			AL	CONF	15
20	COMMON R41(10) R42(10) R43(10) R44(10) R45(10) R46(10) R47(10) R48(10) R49(10) R50(10)			AL	CONF	20
25	COMMON R51(10) R52(10) R53(10) R54(10) R55(10) R56(10) R57(10) R58(10) R59(10) R60(10)			AL	CONF	25
30	COMMON R61(10) R62(10) R63(10) R64(10) R65(10) R66(10) R67(10) R68(10) R69(10) R70(10)			AL	CONF	30
35	COMMON R71(10) R72(10) R73(10) R74(10) R75(10) R76(10) R77(10) R78(10) R79(10) R80(10)			AL	CONF	35
40	COMMON R81(10) R82(10) R83(10) R84(10) R85(10) R86(10) R87(10) R88(10) R89(10) R90(10)			AL	CONF	40
45	COMMON R91(10) R92(10) R93(10) R94(10) R95(10) R96(10) R97(10) R98(10) R99(10) R100(10)			AL	CONF	45
50	ENTRY STATE SOME PRELIMINARY VALUES R1(1)=10 R2(1)=10 R3(1)=10 R4(1)=10 R5(1)=10 R6(1)=10 R7(1)=10 R8(1)=10 R9(1)=10 R10(1)=10			AL	CONF	50
55	R11(1)=10 R12(1)=10 R13(1)=10 R14(1)=10 R15(1)=10 R16(1)=10 R17(1)=10 R18(1)=10 R19(1)=10 R20(1)=10			AL	CONF	55
60	R21(1)=10 R22(1)=10 R23(1)=10 R24(1)=10 R25(1)=10 R26(1)=10 R27(1)=10 R28(1)=10 R29(1)=10 R30(1)=10			AL	CONF	60
65	R31(1)=10 R32(1)=10 R33(1)=10 R34(1)=10 R35(1)=10 R36(1)=10 R37(1)=10 R38(1)=10 R39(1)=10 R40(1)=10			AL	CONF	65
70	R41(1)=10 R42(1)=10 R43(1)=10 R44(1)=10 R45(1)=10 R46(1)=10 R47(1)=10 R48(1)=10 R49(1)=10 R50(1)=10			AL	CONF	70
75	R51(1)=10 R52(1)=10 R53(1)=10 R54(1)=10 R55(1)=10 R56(1)=10 R57(1)=10 R58(1)=10 R59(1)=10 R60(1)=10			AL	CONF	75

310	READ (INTP) U,W,M	AL	270
	READ (INTP) TEMPAND	AL	271
	RETURN	AL	272
315	C DO 201 ZERD ARRAYS	AL	273
	DO 201 W=1,22	AL	274
	DO 201 J=1,NP1	AL	275
	W=0.0	AL	276
	DO 201 I=1,JP1	AL	277
	W=1.0	AL	278
	J=0	AL	279
320	C P=0	AL	280
	DO 451 J=1,NP1	AL	281
	P=U(J)*W	AL	282
	I=0	AL	283
	DO 451 K=1,NP1	AL	284
	I=I+1	AL	285
	DO 451 L=1,NP1	AL	286
	I=I+1	AL	287
325	P=I*W	AL	288
	IF 77.58(I) EQ TO 452	AL	289
	U(I)=J*U(I)-P	AL	290
330	452 W=J*W+U(I)	AL	291
	IF 452(J) EQ TO 451	AL	292
335	451 W=U(I)*W	AL	293
	IF 1LEAR EQ 21 GO TO 295	AL	294
	DO 300 J=1,NP1	AL	295
	DO 300 I=1,NP1	AL	296
	I=I+1	AL	297
	DO 300 J=1,NP1	AL	298
	I=I+1	AL	299
340	C P=J*U(I)*W	AL	300
	DO 295 U(I)=J*U(I)-P	AL	301
	DO 292 I=1,NP1	AL	302
345	U(I)=U(I)-P	AL	303
	U(I)=U(I)+P	AL	304
	DO 292 J=1,NP1	AL	305
350	U(I)=U(I)-P	AL	306
	U(I)=U(I)+P	AL	307
	DO 292 J=1,NP1	AL	308
	U(I)=U(I)-P	AL	309
	U(I)=U(I)+P	AL	310
355	DO 292 J=1,NP1	AL	311
	U(I)=U(I)-P	AL	312
	U(I)=U(I)+P	AL	313
	DO 292 J=1,NP1	AL	314
	U(I)=U(I)-P	AL	315
	U(I)=U(I)+P	AL	316
360	DO 292 J=1,NP1	AL	317
	U(I)=U(I)-P	AL	318
	U(I)=U(I)+P	AL	319
	DO 292 J=1,NP1	AL	320
	U(I)=U(I)-P	AL	321
	U(I)=U(I)+P	AL	322
365	DO 292 J=1,NP1	AL	323
	U(I)=U(I)-P	AL	324
	U(I)=U(I)+P	AL	325
370	DO 292 J=1,NP1	AL	326
	U(I)=U(I)-P	AL	327
	U(I)=U(I)+P	AL	328
375	DO 292 J=1,NP1	AL	329
	U(I)=U(I)-P	AL	330
	U(I)=U(I)+P	AL	331
380	DO 292 J=1,NP1	AL	332
	U(I)=U(I)-P	AL	333
	U(I)=U(I)+P	AL	334
385	DO 292 J=1,NP1	AL	335
	U(I)=U(I)-P	AL	336
	U(I)=U(I)+P	AL	337
390	DO 292 J=1,NP1	AL	338
	U(I)=U(I)-P	AL	339
	U(I)=U(I)+P	AL	340
395	DO 292 J=1,NP1	AL	341
	U(I)=U(I)-P	AL	342
	U(I)=U(I)+P	AL	343
	DO 292 J=1,NP1	AL	344
	U(I)=U(I)-P	AL	345
	U(I)=U(I)+P	AL	346

```

390      FILL(NVKJ,0)
      DYE=(JUL/JEK)*100
      IF (JUL/JEK) 1, 2, 3, 4, 5, 6, 7, 8, 9, 10, 11, 12, 13, 14, 15, 16, 17, 18, 19, 20, 21, 22, 23, 24, 25, 26, 27, 28, 29, 30, 31, 32, 33, 34, 35, 36, 37, 38, 39, 40, 41, 42, 43, 44, 45, 46, 47, 48, 49, 50, 51, 52, 53, 54, 55, 56, 57, 58, 59, 60, 61, 62, 63, 64, 65, 66, 67, 68, 69, 70, 71, 72, 73, 74, 75, 76, 77, 78, 79, 80, 81, 82, 83, 84, 85, 86, 87, 88, 89, 90, 91, 92, 93, 94, 95, 96, 97, 98, 99, 100, 101, 102, 103, 104, 105, 106, 107, 108, 109, 110, 111, 112, 113, 114, 115, 116, 117, 118, 119, 120, 121, 122, 123, 124, 125, 126, 127, 128, 129, 130, 131, 132, 133, 134, 135, 136, 137, 138, 139, 140, 141, 142, 143, 144, 145, 146, 147, 148, 149, 150, 151, 152, 153, 154, 155, 156, 157, 158, 159, 160, 161, 162, 163, 164, 165, 166, 167, 168, 169, 170, 171, 172, 173, 174, 175, 176, 177, 178, 179, 180, 181, 182, 183, 184, 185, 186, 187, 188, 189, 190, 191, 192, 193, 194, 195, 196, 197, 198, 199, 200, 201, 202, 203, 204, 205, 206, 207, 208, 209, 210, 211, 212, 213, 214, 215, 216, 217, 218, 219, 220, 221, 222, 223, 224, 225, 226, 227, 228, 229, 230, 231, 232, 233, 234, 235, 236, 237, 238, 239, 240, 241, 242, 243, 244, 245, 246, 247, 248, 249, 250, 251, 252, 253, 254, 255, 256, 257, 258, 259, 260, 261, 262, 263, 264, 265, 266, 267, 268, 269, 270, 271, 272, 273, 274, 275, 276, 277, 278, 279, 280, 281, 282, 283, 284, 285, 286, 287, 288, 289, 290, 291, 292, 293, 294, 295, 296, 297, 298, 299, 300, 301, 302, 303, 304, 305, 306, 307, 308, 309, 310, 311, 312, 313, 314, 315, 316, 317, 318, 319, 320, 321, 322, 323, 324, 325, 326, 327, 328, 329, 330, 331, 332, 333, 334, 335, 336, 337, 338, 339, 340, 341, 342, 343, 344, 345, 346, 347, 348, 349, 350, 351, 352, 353, 354, 355, 356, 357, 358, 359, 360, 361, 362, 363, 364, 365, 366, 367, 368, 369, 370, 371, 372, 373, 374, 375, 376, 377, 378, 379, 380, 381, 382, 383, 384, 385, 386, 387, 388, 389, 390, 391, 392, 393, 394, 395, 396, 397, 398, 399, 400, 401, 402, 403, 404, 405, 406, 407, 408, 409, 410, 411, 412, 413, 414, 415, 416, 417, 418, 419, 420, 421, 422, 423, 424, 425, 426, 427, 428, 429, 430, 431, 432, 433, 434, 435, 436, 437, 438, 439, 440, 441, 442, 443, 444, 445, 446, 447, 448, 449, 450, 451, 452, 453, 454, 455, 456, 457, 458, 459, 460, 461, 462, 463, 464, 465, 466, 467, 468, 469, 470, 471, 472, 473, 474, 475, 476, 477, 478, 479, 480, 481, 482, 483, 484, 485, 486, 487, 488, 489, 490, 491, 492, 493, 494, 495, 496, 497, 498, 499, 500, 501, 502, 503, 504, 505, 506, 507, 508, 509, 510, 511, 512, 513, 514, 515, 516, 517, 518, 519, 520, 521, 522, 523, 524, 525, 526, 527, 528, 529, 530, 531, 532, 533, 534, 535, 536, 537, 538, 539, 540, 541, 542, 543, 544, 545, 546, 547, 548, 549, 550, 551, 552, 553, 554, 555, 556, 557, 558, 559, 560, 561, 562, 563, 564, 565, 566, 567, 568, 569, 570, 571, 572, 573, 574, 575, 576, 577, 578, 579, 580, 581, 582, 583, 584, 585, 586, 587, 588, 589, 590, 591, 592, 593, 594, 595, 596, 597, 598, 599, 600, 601, 602, 603, 604, 605, 606, 607, 608, 609, 610, 611, 612, 613, 614, 615, 616, 617, 618, 619, 620, 621, 622, 623, 624, 625, 626, 627, 628, 629, 630, 631, 632, 633, 634, 635, 636, 637, 638, 639, 640, 641, 642, 643, 644, 645, 646, 647, 648, 649, 650, 651, 652, 653, 654, 655, 656, 657, 658, 659, 660, 661, 662, 663, 664, 665, 666, 667, 668, 669, 670, 671, 672, 673, 674, 675, 676, 677, 678, 679, 680, 681, 682, 683, 684, 685, 686, 687, 688, 689, 690, 691, 692, 693, 694, 695, 696, 697, 698, 699, 700, 701, 702, 703, 704, 705, 706, 707, 708, 709, 710, 711, 712, 713, 714, 715, 716, 717, 718, 719, 720, 721, 722, 723, 724, 725, 726, 727, 728, 729, 730, 731, 732, 733, 734, 735, 736, 737, 738, 739, 740, 741, 742, 743, 744, 745, 746, 747, 748, 749, 750, 751, 752, 753, 754, 755, 756, 757, 758, 759, 760, 761, 762, 763, 764, 765, 766, 767, 768, 769, 770, 771, 772, 773, 774, 775, 776, 777, 778, 779, 780, 781, 782, 783, 784, 785, 786, 787, 788, 789, 790, 791, 792, 793, 794, 795, 796, 797, 798, 799, 800, 801, 802, 803, 804, 805, 806, 807, 808, 809, 810, 811, 812, 813, 814, 815, 816, 817, 818, 819, 820, 821, 822, 823, 824, 825, 826, 827, 828, 829, 830, 831, 832, 833, 834, 835, 836, 837, 838, 839, 840, 841, 842, 843, 844, 845, 846, 847, 848, 849, 850, 851, 852, 853, 854, 855, 856, 857, 858, 859, 860, 861, 862, 863, 864, 865, 866, 867, 868, 869, 870, 871, 872, 873, 874, 875, 876, 877, 878, 879, 880, 881, 882, 883, 884, 885, 886, 887, 888, 889, 890, 891, 892, 893, 894, 895, 896, 897, 898, 899, 900, 901, 902, 903, 904, 905, 906, 907, 908, 909, 910, 911, 912, 913, 914, 915, 916, 917, 918, 919, 920, 921, 922, 923, 924, 925, 926, 927, 928, 929, 930, 931, 932, 933, 934, 935, 936, 937, 938, 939, 940, 941, 942, 943, 944, 945, 946, 947, 948, 949, 950, 951, 952, 953, 954, 955, 956, 957, 958, 959, 960, 961, 962, 963, 964, 965, 966, 967, 968, 969, 970, 971, 972, 973, 974, 975, 976, 977, 978, 979, 980, 981, 982, 983, 984, 985, 986, 987, 988, 989, 990, 991, 992, 993, 994, 995, 996, 997, 998, 999, 1000

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SMARNTIME ALLM00 73774 OPT=0 TRACE FTN 6,50430 09/26/79 16.09.65

155	3001 CAR(1,J,43)-AMOUNTPLUS/PREFINVT/(ALOG(EOTPLUS)/AKOPJAT(VN))		AL	623
	GO TO 3007		AL	624
	CALL(J,1,14)		AL	625
160	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3006		AL	627
	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3006		AL	628
	CALL(J,43)-AMOUNTPLUS		AL	629
165	GO TO 3007		AL	630
	CALL(J,43)-AMOUNTPLUS		AL	631
	CALL(J,43)-AMOUNTPLUS		AL	632
	CALL(J,43)-AMOUNTPLUS		AL	633
	CALL(J,43)-AMOUNTPLUS		AL	634
	CALL(J,43)-AMOUNTPLUS		AL	635
	CALL(J,43)-AMOUNTPLUS		AL	636
	CALL(J,43)-AMOUNTPLUS		AL	637
	CALL(J,43)-AMOUNTPLUS		AL	638
	CALL(J,43)-AMOUNTPLUS		AL	639
170	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3006		AL	640
	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3006		AL	641
175	CALL(J,43)-AMOUNTPLUS		AL	642
	CALL(J,43)-AMOUNTPLUS		AL	643
	CALL(J,43)-AMOUNTPLUS		AL	644
	CALL(J,43)-AMOUNTPLUS		AL	645
	CALL(J,43)-AMOUNTPLUS		AL	646
	CALL(J,43)-AMOUNTPLUS		AL	647
180	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3012		AL	648
	IF ((EMN(FN)) OR (EMJ(FN))) GO TO 3012		AL	649
185	CALL(J,43)-AMOUNTPLUS		AL	650
	CALL(J,43)-AMOUNTPLUS		AL	651
	CALL(J,43)-AMOUNTPLUS		AL	652
	CALL(J,43)-AMOUNTPLUS		AL	653
	CALL(J,43)-AMOUNTPLUS		AL	654
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190	CALL(J,43)-AMOUNTPLUS		AL	657
	CALL(J,43)-AMOUNTPLUS		AL	658
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195	CALL(J,43)-AMOUNTPLUS		AL	662
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200	CALL(J,43)-AMOUNTPLUS		AL	668
	CALL(J,43)-AMOUNTPLUS		AL	669
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205	CALL(J,43)-AMOUNTPLUS		AL	674
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	CALL(J,43)-AMOUNTPLUS		AL	698
	CALL(J,43)-AMOUNTPLUS		AL	699
	CALL(J,43)-AMOUNTPLUS		AL	700


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628 66.4740158E01
    LFC=AK(1)-1
    927 SP(1)SPLI-JJ-EVAP(LPCI
    C-----
    980 IF (OPT=NO) GO TO 910
    K-DIRECTION RADIATION
    IF (OPT=NO) GO TO 910
    I=1
    DO 1001 I=1,6
      KEM(1)=E1
      IF (KEM(1)=0) GO TO 902
      SU1(1)=SUI(1)-EM
      SGM(1)=SIGNA(TEMP004)
    1001 CONTINUE
    SU1(1)=SUI(1)-RADI
    SGM(1)=SIGNA(TEMP011)
    I=2
    DO 1002 I=2,6
      KEM(1)=KEM(1)+LT.JMOJ GO TO 1002
      SU1(1)=SUI(1)-EM
      SGM(1)=SIGNA(TEMP004)
    1002 CONTINUE
    SUI(1)=SUI(1)-RADI
    SGM(1)=SIGNA(TEMP004)
    901 CONTINUE
    COME HERE FOR SLOT MODIFICATIONS
    IF (SU1(1)=0) GO TO 963
    I=1
    DO 1003 I=1,6
      KEM(1)=E1
      SU1(1)=SUI(1)-EM
      SGM(1)=SIGNA(TEMP004)
    1003 CONTINUE
    SU1(1)=SUI(1)-EM
    SGM(1)=SIGNA(TEMP004)
    743 SUI(1)=SUI(1)-EM
    703 CONTINUE
    910 IF (ELRY) GO TO 920
    J=JMOD(1)
    IF (J=1) GO TO 904
    IF (I=1) KEM(1)=E0+2 OR (KEM(1)=E0+3) GO TO 904
    SU1(1)=SUI(1)-EM
    SGM(1)=SIGNA(TEMP004)
    GO TO 905
    904 SUI(1)=SUI(1)-RADI
    905 IF (SU1(1)=0) GO TO 911
    1011 IF (I=1) KEM(1)=E0+1 OR (KEM(1)=E0+3) GO TO 906
    SU1(1)=SUI(1)-EM
    SGM(1)=SIGNA(TEMP004)
    906 SUI(1)=SUI(1)-RADI
    911 CONTINUE
    IF (SU1(1)=0) GO TO 912
    SUI(1)=SUI(1)-EM
    SGM(1)=SIGNA(TEMP004)
    I=J-1
    IF (I=0) SUI(1)=E0+1
    IF (I=0) SU1(1)=SUI(1)-EM
    744 SUI(1)=SUI(1)-EM
    912 CONTINUE
    920 SGM(1)=SIGNA(TEMP012)
    C-----
    600 Entry 3000Z
    IF (I=0) GO TO 1106
    I=1
    DO 1107 I=1,6
      KEM(1)=E1
      SU1(1)=SUI(1)-EM
      SGM(1)=SIGNA(TEMP004)
    1107 CONTINUE
    SUI(1)=SUI(1)-RADI
    SGM(1)=SIGNA(TEMP004)
    1121 SUI(1)=SUI(1)-EM
    1122 CONTINUE
    1123 SUI(1)=SUI(1)-EM
    1124 SUI(1)=SUI(1)-EM
    1125 SUI(1)=SUI(1)-EM
    1126 SUI(1)=SUI(1)-EM
    1127 SUI(1)=SUI(1)-EM
    1128 SUI(1)=SUI(1)-EM
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SUBROUTINE ALLMOD 73/74 OPT=0 TRACE

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695      IF (COUNT(JL1)-ME.JPLANE) GO TO 1102
        IC=1104
        I=1104
        SUI=41-01
        SPLANE=1.E30
        CONTINUE
        1104      IF (COUNT(JPLANE)-4)
        1106      DO 921 AC=1,SPLE=3*KRZ1
        1108      LPA=1+COUNT(JPLANE)-KRZ1
        1110      CAMODL=50*(GANT(JPLANE,Z)+GANT(JPLANE,N))/JPLANE/(ZD1*(MP1))
        1112      CAMODL=ZD1*(Z1)
        1114      SUI=21-SUI
        1116      SPLANE=SPLE
        1118      SPLANE=SPLE
        1120      CONTINUE
        1122      RETURN
        1124      END
    
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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM
 72 1 CONTROL VARIABLE IN COMMON OR EQUIVALENTS: OPTIMIZATION MAY BE INTERRUPTED.
 73 1 CONTROL VARIABLE IN COMMON OR EQUIVALENTS: OPTIMIZATION MAY BE INTERRUPTED.
 74 1 CONTROL VARIABLE IN COMMON OR EQUIVALENTS: OPTIMIZATION MAY BE INTERRUPTED.
 75 1 CONTROL VARIABLE IN COMMON OR EQUIVALENTS: OPTIMIZATION MAY BE INTERRUPTED.

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1 SUBROUTINE QUIT(C)
2 COMMON /S/ I, J, K, L, M, N, O, P, Q, R, S, T, U, V, W, X, Y, Z, AA, AB, AC, AD, AE, AF, AG, AH, AI, AJ, AK, AL, AM, AN, AO, AP, AQ, AR, AS, AT, AU, AV, AW, AX, AY, AZ, BA, BB, BC, BD, BE, BF, BG, BH, BI, BJ, BK, BL, BM, BN, BO, BP, BQ, BR, BS, BT, BU, BV, BW, BX, BY, BZ, CA, CB, CC, CD, CE, CF, CG, CH, CI, CJ, CK, CL, CM, CN, CO, CP, CQ, CR, CS, CT, CU, CV, CW, CX, CY, CZ, DA, DB, DC, DD, DE, DF, DG, DH, DI, DJ, DK, DL, DM, DN, DO, DP, DQ, DR, DS, DT, DU, DV, DW, DX, DY, DZ, EA, EB, EC, ED, EE, EF, EG, EH, EI, EJ, EK, EL, EM, EN, EO, EP, EQ, ER, ES, ET, EU, EV, EW, EX, EY, EZ, FA, FB, FC, FD, FE, FF, FG, FH, FI, FJ, FK, FL, FM, FN, FO, FP, FQ, FR, FS, FT, FU, FV, FW, FX, FY, FZ, GA, GB, GC, GD, GE, GF, GG, GH, GI, GJ, GK, GL, GM, GN, GO, GP, GQ, GR, GS, GT, GU, GV, GW, GX, GY, GZ, HA, HB, HC, HD, HE, HF, HG, HH, HI, HJ, HK, HL, HM, HN, HO, HP, HQ, HR, HS, HT, HU, HV, HW, HX, HY, HZ, IA, IB, IC, ID, IE, IF, IG, IH, II, IJ, IK, IL, IM, IN, IO, IP, IQ, IR, IS, IT, IU, IV, IW, IX, IY, IZ, JA, JB, JC, JD, JE, JF, JG, JH, JI, JJ, JK, JL, JM, JN, JO, JP, JQ, JR, JS, JT, JU, JV, JW, JX, JY, JZ, KA, KB, KC, KD, KE, KF, KG, KH, KI, KJ, KK, KL, KM, KN, KO, KP, KQ, KR, KS, KT, KU, KV, KW, KX, KY, KZ, LA, LB, LC, LD, LE, LF, LG, LH, LI, LJ, LK, LL, LM, LN, LO, LP, LQ, LR, LS, LT, LU, LV, LW, LX, LY, LZ, MA, MB, MC, MD, ME, MF, MG, MH, MI, MJ, MK, ML, MM, MN, MO, MP, MQ, MR, MS, MT, MU, MV, MW, MX, MY, MZ, NA, NB, NC, ND, NE, NF, NG, NH, NI, NJ, NK, NL, NM, NN, NO, NP, NQ, NR, NS, NT, NU, NV, NW, NX, NY, NZ, OA, OB, OC, OD, OE, OF, OG, OH, OI, OJ, OK, OL, OM, ON, OO, OP, OQ, OR, OS, OT, OU, OV, OW, OX, OY, OZ, PA, PB, PC, PD, PE, PF, PG, PH, PI, PJ, PK, PL, PM, PN, PO, PP, PQ, PR, PS, PT, PU, PV, PW, PX, PY, PZ, QA, QB, QC, QD, QE, QF, QG, QH, QI, QJ, QK, QL, QM, QN, QO, QP, QQ, QR, QS, QT, QU, QV, QW, QX, QY, QZ, RA, RB, RC, RD, RE, RF, RG, RH, RI, RJ, RK, RL, RM, RN, RO, RP, RQ, RR, RS, RT, RU, RV, RW, RX, RY, RZ, SA, SB, SC, SD, SE, SF, SG, SH, SI, SJ, SK, SL, SM, SN, SO, SP, SQ, SR, SS, ST, SU, SV, SW, SX, SY, SZ, TA, TB, TC, TD, TE, TF, TG, TH, TI, TJ, TK, TL, TM, TN, TO, TP, TQ, TR, TS, TT, TU, TV, TW, TX, TY, TZ, UA, UB, UC, UD, UE, UF, UG, UH, UI, UJ, UK, UL, UM, UN, UO, UP, UQ, UR, US, UT, UY, UV, UW, UX, UY, UZ, VA, VB, VC, VD, VE, VF, VG, VH, VI, VJ, VK, VL, VM, VN, VO, VP, VQ, VR, VS, VT, VU, VV, VW, VX, VY, VZ, WA, WB, WC, WD, WE, WF, WG, WH, WI, WJ, WK, WL, WM, WN, WO, WP, WQ, WR, WS, WT, WU, WV, WW, WX, WY, WZ, XA, XB, XC, XD, XE, XF, XG, XH, XI, XJ, XK, XL, XM, XN, XO, XP, XQ, XR, XS, XT, XU, XV, XW, XX, XY, XZ, YA, YB, YC, YD, YE, YF, YG, YH, YI, YJ, YK, YL, YM, YN, YO, YP, YQ, YR, YS, YT, YU, YV, YW, YX, YY, YZ, ZA, ZB, ZC, ZD, ZE, ZF, ZG, ZH, ZI, ZJ, ZK, ZL, ZM, ZN, ZO, ZP, ZQ, ZR, ZS, ZT, ZU, ZV, ZW, ZX, ZY, ZZ
3 CALL QUIT(C)
4 RETURN
5 END

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80      I COM(F,LP,NM,20)
      CALL FPRINT (MVHZ0,MVHZ,16)
      IF (MPCZ,LE,0) GO TO 35
      DO 37 I=1,MPI
      KJM=KMK(I)*JMJ
      DD 37 I=1,LP1
      LP=KJM
      37 FLP=MVFOUKI=0.
      DD 30 I=1,N
      KJM=KMK(I)*JMJ
      KK=(K-2)*(MI-2)*(NJ-2)+(J-2)*(MI-2)
      DD 30 I=1,2*L
      LCM=KK*(I-1)
      39 FLP=MVFOUKI=CMVAP(LPC)
      CALL FPRINT (MVFOUK,MVFOU,22)
C-----
      CONTINUE OV IS ENHALPY, DV IS FAV
      CALL FPRINT (MVHNVAV,10)
C-----
      READ HERE IS EX. @ IS Py, M IS FZ
      READ FPRINT (MVK,MVZ,12)
      CALL FPRINT (II,II,20)
      MV=4
      DD 50 I=1,MPI
      KJM=KMK(I)*JMJ
      DD 50 I=1,LP1
      LP=KJM
      56 FLP=ZKQ
      CALL FPRINT (7,7,21)
      READ (M,II,1,2)
      46 READ (M,II) IS PWL, P IS MFLU, DV IS MCO
      16 CONTINUE
      19 WRITE (6,10)
      19 FORMAT (10H,ANSWER IS,ANSWER,7F,4V5,5SUM,6K,5M,8E11,5X,
      2  10H,END OF PRINTING)
      20 CONTINUE
      RETURN
      END
  
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CARD NR. SEVERITY DETAILS           DIAGNOSIS OF PROBLEM
55      I      CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE IMHIBITED.
56      I      CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE IMHIBITED.
57      I      CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE IMHIBITED.
58      I      CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE IMHIBITED.
107     I      CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE IMHIBITED.
  
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SUBROUTINE AUX
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COMMON D(1500),DU(1500),DV(1500),DW(1500),DX(1500),DY(1500),DZ(1500),
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80      CALL DWRITE(LINE)
81      CALL DWRITE('*****')
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148     CALL DWRITE('*****')
149     CALL DWRITE('*****')
150     CALL DWRITE('*****')

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235 CARLPLVISC(LP)/PREMVS)
    COMT)
    CARLPLVISC(LP)/PREFINS)
240 3700 CONTINUE
    3004 CALL CASDS
    CONTINUE
    C . . . . .
    ENTRY SOURCE
    CAUTION NOT ALL SOURCE TERMS ARE VALID - AT BOUNDARY MODES -
    CHECK AND VERIFY ACCORDINGLY TO SOURCE
    GO TO 100,200,300,400,500
245 C-----SOURCE TERMS FOR U-VELOCITY -----
    100 70 101 102
    101 101 101 101
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TIME	ADDRESS	OPERATION	DATA
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LINE	FUNCTION GASP	DATE TRACE	FIM 6.4-430	08/24/78 14.06.46	PAGE 1
1	FUNCTION GASP (17,12)				
2	DIMENSIONAL (17,12) (12)				
3	SPECIES TRAJ - FUEL (C2,C2,C2,M2O,2M,3,M,M2,4,M2,M2)				
4	DATA (14,11,12,13,14,15,16,17,18,19,20,21,22,23,24,25,26,27,28,29,30,31,32,33,34,35,36,37,38,39,40,41,42,43,44,45,46,47,48,49,50,51,52,53,54,55,56,57,58,59,60,61,62,63,64,65,66,67,68,69,70,71,72,73,74,75,76,77,78,79)				
5	1	0.255339E+02	0.667053E+01	0.2295723E+04	5.4949416E+00
6	2	0.242001E+01	0.338711E+00	0.12382571E+05	0.22741325E+07
7	3	0.155056E+01	0.489148E+00	0.57833021E+00	0.10364577E+00
8	4	0.233333E+01	0.425280E+00	0.31679154E+01	0.35201558E+10
9	5	0.277777E+01	0.361324E+00	0.1662228E+05	0.13228008E+00
10	6	0.277777E+01	0.211137E+00	0.21250794E+01	0.13228008E+00
11	7	0.442122E+01	0.290182E+00	0.63183671E+01	0.12730794E+10
12	8	0.233333E+01	0.390700E+00	0.54288735E+01	
13	9	0.155056E+01	0.127440E+00	0.31028038E+00	0.55112974E+11
14	10	0.255339E+01	0.292306E+00	0.40283540E+01	0.
15	11	0.233333E+01	0.257162E+00	0.4901703E+00	0.
16	12	0.155056E+01	0.111465E+00	0.2664420E+07	0.3490973E+00
17	13	0.233333E+01	0.104194E+00	0.14457311E+01	0.1879624E+10
18	14	0.233333E+01	0.111465E+00	0.44437311E+01	0.95919332E+10
19	15	0.233333E+01	0.138228E+00	0.28899318E+06	0.99837393E+10
20	16	0.442122E+01	0.382323E+00	0.57498129E+01	0.
21	17	0.233333E+01	0.153436E+00	0.2723377E+00	0.
22	18	0.233333E+01	0.405618E+00	0.61633168E+01	0.
23	19	0.233333E+01	0.734314E+00	0.51144255E+04	0.346982140E+07
24	20	0.233333E+01	0.359149E+00	0.	0.2021851E+00
25	21	0.233333E+01	0.272327E+00	0.4601097E+00	0.
26	22	0.233333E+01	0.172797E+00	0.2021851E+00	0.
27	23	0.233333E+01	0.143901E+00	0.20319674E+00	0.
28	24	0.233333E+01	0.143901E+00	0.7055331E+01	0.
29	25	0.233333E+01	0.172797E+00	0.7055331E+01	0.6743137E+00
30	26	0.233333E+01	0.172797E+00	0.4307278E+01	0.
31	27	0.233333E+01	0.104194E+00	0.21250794E+01	0.
32	28	0.233333E+01	0.104194E+00	0.13250794E+01	0.
33	29	0.233333E+01	0.3541232E+00	0.4937000E+00	0.
34	30	0.233333E+01	0.172797E+00	0.4232326E+00	0.
35	31	0.233333E+01	0.104194E+00	0.292306E+00	0.
36	32	0.233333E+01	0.104194E+00	0.292306E+00	0.
37	33	0.233333E+01	0.104194E+00	0.292306E+00	0.
38	34	0.233333E+01	0.104194E+00	0.292306E+00	0.
39	35	0.233333E+01	0.104194E+00	0.292306E+00	0.
40	36	0.233333E+01	0.104194E+00	0.292306E+00	0.
41	37	0.233333E+01	0.104194E+00	0.292306E+00	0.
42	38	0.233333E+01	0.104194E+00	0.292306E+00	0.
43	39	0.233333E+01	0.104194E+00	0.292306E+00	0.
44	40	0.233333E+01	0.104194E+00	0.292306E+00	0.
45	41	0.233333E+01	0.104194E+00	0.292306E+00	0.
46	42	0.233333E+01	0.104194E+00	0.292306E+00	0.
47	43	0.233333E+01	0.104194E+00	0.292306E+00	0.
48	44	0.233333E+01	0.104194E+00	0.292306E+00	0.
49	45	0.233333E+01	0.104194E+00	0.292306E+00	0.
50	46	0.233333E+01	0.104194E+00	0.292306E+00	0.
51	47	0.233333E+01	0.104194E+00	0.292306E+00	0.
52	48	0.233333E+01	0.104194E+00	0.292306E+00	0.
53	49	0.233333E+01	0.104194E+00	0.292306E+00	0.
54	50	0.233333E+01	0.104194E+00	0.292306E+00	0.
55	51	0.233333E+01	0.104194E+00	0.292306E+00	0.
56	52	0.233333E+01	0.104194E+00	0.292306E+00	0.
57	53	0.233333E+01	0.104194E+00	0.292306E+00	0.
58	54	0.233333E+01	0.104194E+00	0.292306E+00	0.
59	55	0.233333E+01	0.104194E+00	0.292306E+00	0.
60	56	0.233333E+01	0.104194E+00	0.292306E+00	0.
61	57	0.233333E+01	0.104194E+00	0.292306E+00	0.
62	58	0.233333E+01	0.104194E+00	0.292306E+00	0.
63	59	0.233333E+01	0.104194E+00	0.292306E+00	0.
64	60	0.233333E+01	0.104194E+00	0.292306E+00	0.
65	61	0.233333E+01	0.104194E+00	0.292306E+00	0.
66	62	0.233333E+01	0.104194E+00	0.292306E+00	0.
67	63	0.233333E+01	0.104194E+00	0.292306E+00	0.
68	64	0.233333E+01	0.104194E+00	0.292306E+00	0.
69	65	0.233333E+01	0.104194E+00	0.292306E+00	0.
70	66	0.233333E+01	0.104194E+00	0.292306E+00	0.
71	67	0.233333E+01	0.104194E+00	0.292306E+00	0.
72	68	0.233333E+01	0.104194E+00	0.292306E+00	0.
73	69	0.233333E+01	0.104194E+00	0.292306E+00	0.
74	70	0.233333E+01	0.104194E+00	0.292306E+00	0.
75	71	0.233333E+01	0.104194E+00	0.292306E+00	0.
76	72	0.233333E+01	0.104194E+00	0.292306E+00	0.
77	73	0.233333E+01	0.104194E+00	0.292306E+00	0.
78	74	0.233333E+01	0.104194E+00	0.292306E+00	0.
79	75	0.233333E+01	0.104194E+00	0.292306E+00	0.

```

00 IF (I.EQ.1) T=AMIN(I,I)
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
05 RETURN
   PRINT *
   IF (I.EQ.1) T=AMIN(I,I)
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
   T=AM+AM/2-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
09 RETURN
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
100 RETURN
   PRINT *
   IF (I.EQ.1) T=AMIN(I,I)
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
   T=AM+AM/2-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
105 RETURN
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
110 RETURN
   PRINT *
   IF (I.EQ.1) T=AMIN(I,I)
   AM=AM+I-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
   T=AM+AM/2-I*(I+AM)/2-I*(I+AM)/3-I*(I+AM)/6-I*(I+AM)/15-I*(I+AM)/30
   END

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1 SUBROUTINE AURAD
COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
5 A=1.0, B=1.0, C=1.0, D=1.0, E=1.0, F=1.0, G=1.0, H=1.0, I=1.0, J=1.0
10 DIMENSION A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
15 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
20 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
25 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
30 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
35 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
40 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
45 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
50 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
55 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
60 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
65 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
70 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)
75 COMMON /AURAD/ A(1000), B(1000), C(1000), D(1000), E(1000), F(1000), G(1000), H(1000), I(1000), J(1000)

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1      SUBROUTINE SPRAY
2      SUBROUTINE SPRAY
3      SUBROUTINE FOR ATRIP CODE WHICH CALCULATES THE TRAJECTORY
4      AND SADDLE RATES FOR A FUEL NOZZLE SPRAY. WRITTEN NOV. 1976
5      BY.....
6      C.....
7      C.....
8      C.....
9      C.....
10     C.....
11     C.....
12     C.....
13     C.....
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75     C.....

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155 GO TO 11
156 CONTINUE
157 LOC=K(AM)-GT.ZI(KS) GO TO 15
158 CONTINUE
159 CONTINUE
160 CONTINUE
161 CONTINUE
162 CONTINUE
163 CONTINUE
164 CONTINUE
165 CONTINUE
166 CONTINUE
167 CONTINUE
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218 CONTINUE
219 CONTINUE
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223 CONTINUE
224 CONTINUE
225 CONTINUE
226 CONTINUE
227 CONTINUE
228 CONTINUE
229 CONTINUE
230 CONTINUE

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C-----
199 TRANSFER NUMBER
200 IF (LTIME)-STIME) 210=205-201
201 SET AMPL=1(CPI)-(YST)-(E)/MEVAP*0.1
202 GO TO 210
203 INITIALIZE
210 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
211 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
212 CONTINUE
213 GO TO 210
220 IF (FEVAP-I)-GO TO 220
221 FEVAP=FEVAP+FEVAP*1.
222 CONTINUE
223 GO TO 220
230 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
231 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
232 CONTINUE
233 GO TO 230
240 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
241 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
242 CONTINUE
243 GO TO 240
250 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
251 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
252 CONTINUE
253 GO TO 250
260 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
261 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
262 CONTINUE
263 GO TO 260
270 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
271 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
272 CONTINUE
273 GO TO 270
280 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
281 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
282 CONTINUE
283 GO TO 280
290 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
291 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
292 CONTINUE
293 GO TO 290
300 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
301 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
302 CONTINUE
303 GO TO 300
310 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
311 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
312 CONTINUE
313 GO TO 310
320 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
321 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
322 CONTINUE
323 GO TO 320
330 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
331 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
332 CONTINUE
333 GO TO 330
340 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
341 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
342 CONTINUE
343 GO TO 340
350 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
351 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
352 CONTINUE
353 GO TO 350
360 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
361 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
362 CONTINUE
363 GO TO 360
370 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
371 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
372 CONTINUE
373 GO TO 370
380 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
381 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
382 CONTINUE
383 GO TO 380
390 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
391 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
392 CONTINUE
393 GO TO 390
400 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
401 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
402 CONTINUE
403 GO TO 400
410 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
411 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
412 CONTINUE
413 GO TO 410
420 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
421 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
422 CONTINUE
423 GO TO 420
430 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
431 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
432 CONTINUE
433 GO TO 430
440 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
441 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
442 CONTINUE
443 GO TO 440
450 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
451 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
452 CONTINUE
453 GO TO 450
460 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
461 E=C*(1+(LTIME-STIME)/AMPL*(CPI)-(YST)-(E)/MEVAP
462 CONTINUE
463 GO TO 460

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3P 42 428

RETURN
END

CARD NO.	SEVERITY	DETAILS	DIAGNOSIS OF PROBLEM
27			CONTROL VARIABLE IN COMMON
317			OR EQUIVALENTS
376			OR EQUIVALENTS
432			OR EQUIVALENTS
443			OR EQUIVALENTS

CONTROL VARIABLE IN COMMON
 CONTROL VARIABLE IN COMMON
 CONTROL VARIABLE IN COMMON
 AN IN STATEMENT
 CONTROL VARIABLE IN COMMON
 CONTROL VARIABLE IN COMMON

OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS

OPTIMIZATION
 OPTIMIZATION
 OPTIMIZATION
 OPTIMIZATION
 OPTIMIZATION

MAY BE INITIALIZED
 MAY BE INITIALIZED
 MAY BE INITIALIZED
 MAY BE INITIALIZED
 MAY BE INITIALIZED

OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS
 OR EQUIVALENTS

STATEMENT

```

1  SYNTAX ERROR (L,IX,II,NTAB)
2  SYNTAX ERROR (L,IX,II,NTAB)
3  SYNTAX ERROR (L,IX,II,NTAB)
4  SYNTAX ERROR (L,IX,II,NTAB)
5  SYNTAX ERROR (L,IX,II,NTAB)
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7  SYNTAX ERROR (L,IX,II,NTAB)
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61 SYNTAX ERROR (L,IX,II,NTAB)
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64 SYNTAX ERROR (L,IX,II,NTAB)
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86 SYNTAX ERROR (L,IX,II,NTAB)
87 SYNTAX ERROR (L,IX,II,NTAB)
88 SYNTAX ERROR (L,IX,II,NTAB)
89 SYNTAX ERROR (L,IX,II,NTAB)
90 SYNTAX ERROR (L,IX,II,NTAB)
91 SYNTAX ERROR (L,IX,II,NTAB)
92 SYNTAX ERROR (L,IX,II,NTAB)
93 SYNTAX ERROR (L,IX,II,NTAB)
94 SYNTAX ERROR (L,IX,II,NTAB)
95 SYNTAX ERROR (L,IX,II,NTAB)
96 SYNTAX ERROR (L,IX,II,NTAB)
97 SYNTAX ERROR (L,IX,II,NTAB)
98 SYNTAX ERROR (L,IX,II,NTAB)
99 SYNTAX ERROR (L,IX,II,NTAB)
100 SYNTAX ERROR (L,IX,II,NTAB)

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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM
7 I XX ARRAY REFERENCE OUTSIDE DIMENSION BOUNDS.

```

1  SUBROUTINE TRACE
2  DIMENSION TRIDE(100), TRID(100), TRIDC(100), TRIDC2(100), TRIDC3(100), TRIDC4(100), TRIDC5(100), TRIDC6(100), TRIDC7(100), TRIDC8(100), TRIDC9(100), TRIDC10(100), TRIDC11(100), TRIDC12(100), TRIDC13(100), TRIDC14(100), TRIDC15(100), TRIDC16(100), TRIDC17(100), TRIDC18(100), TRIDC19(100), TRIDC20(100), TRIDC21(100), TRIDC22(100), TRIDC23(100), TRIDC24(100), TRIDC25(100), TRIDC26(100), TRIDC27(100), TRIDC28(100), TRIDC29(100), TRIDC30(100), TRIDC31(100), TRIDC32(100), TRIDC33(100), TRIDC34(100), TRIDC35(100), TRIDC36(100), TRIDC37(100), TRIDC38(100), TRIDC39(100), TRIDC40(100), TRIDC41(100), TRIDC42(100), TRIDC43(100), TRIDC44(100), TRIDC45(100), TRIDC46(100), TRIDC47(100), TRIDC48(100), TRIDC49(100), TRIDC50(100), TRIDC51(100), TRIDC52(100), TRIDC53(100), TRIDC54(100), TRIDC55(100), TRIDC56(100), TRIDC57(100), TRIDC58(100), TRIDC59(100), TRIDC60(100), TRIDC61(100), TRIDC62(100), TRIDC63(100), TRIDC64(100), TRIDC65(100), TRIDC66(100), TRIDC67(100), TRIDC68(100), TRIDC69(100), TRIDC70(100), TRIDC71(100), TRIDC72(100), TRIDC73(100), TRIDC74(100), TRIDC75(100), TRIDC76(100), TRIDC77(100), TRIDC78(100), TRIDC79(100), TRIDC80(100), TRIDC81(100), TRIDC82(100), TRIDC83(100), TRIDC84(100), TRIDC85(100), TRIDC86(100), TRIDC87(100), TRIDC88(100), TRIDC89(100), TRIDC90(100), TRIDC91(100), TRIDC92(100), TRIDC93(100), TRIDC94(100), TRIDC95(100), TRIDC96(100), TRIDC97(100), TRIDC98(100), TRIDC99(100), TRIDC100(100)
3  DO 100 I=1,100
4  TRIDE(I)=0
5  TRID(I)=0
6  TRIDC(I,1)=0
7  TRIDC(I,2)=0
8  TRIDC(I,3)=0
9  TRIDC(I,4)=0
10 TRIDC(I,5)=0
11 TRIDC(I,6)=0
12 TRIDC(I,7)=0
13 TRIDC(I,8)=0
14 TRIDC(I,9)=0
15 TRIDC(I,10)=0
16 TRIDC(I,11)=0
17 TRIDC(I,12)=0
18 TRIDC(I,13)=0
19 TRIDC(I,14)=0
20 TRIDC(I,15)=0
21 TRIDC(I,16)=0
22 TRIDC(I,17)=0
23 TRIDC(I,18)=0
24 TRIDC(I,19)=0
25 TRIDC(I,20)=0
26 TRIDC(I,21)=0
27 TRIDC(I,22)=0
28 TRIDC(I,23)=0
29 TRIDC(I,24)=0
30 TRIDC(I,25)=0
31 TRIDC(I,26)=0
32 TRIDC(I,27)=0
33 TRIDC(I,28)=0
34 TRIDC(I,29)=0
35 TRIDC(I,30)=0
36 TRIDC(I,31)=0
37 TRIDC(I,32)=0
38 TRIDC(I,33)=0
39 TRIDC(I,34)=0
40 TRIDC(I,35)=0
41 TRIDC(I,36)=0
42 TRIDC(I,37)=0
43 TRIDC(I,38)=0
44 TRIDC(I,39)=0
45 TRIDC(I,40)=0
46 TRIDC(I,41)=0
47 TRIDC(I,42)=0
48 TRIDC(I,43)=0
49 TRIDC(I,44)=0
50 TRIDC(I,45)=0
51 TRIDC(I,46)=0
52 TRIDC(I,47)=0
53 TRIDC(I,48)=0
54 TRIDC(I,49)=0
55 TRIDC(I,50)=0
56 TRIDC(I,51)=0
57 TRIDC(I,52)=0
58 TRIDC(I,53)=0
59 TRIDC(I,54)=0
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67 TRIDC(I,62)=0
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72 TRIDC(I,67)=0
73 TRIDC(I,68)=0
74 TRIDC(I,69)=0
75 TRIDC(I,70)=0
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78 TRIDC(I,73)=0
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81 TRIDC(I,76)=0
82 TRIDC(I,77)=0
83 TRIDC(I,78)=0
84 TRIDC(I,79)=0
85 TRIDC(I,80)=0
86 TRIDC(I,81)=0
87 TRIDC(I,82)=0
88 TRIDC(I,83)=0
89 TRIDC(I,84)=0
90 TRIDC(I,85)=0
91 TRIDC(I,86)=0
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93 TRIDC(I,88)=0
94 TRIDC(I,89)=0
95 TRIDC(I,90)=0
96 TRIDC(I,91)=0
97 TRIDC(I,92)=0
98 TRIDC(I,93)=0
99 TRIDC(I,94)=0
100 TRIDC(I,95)=0
101 TRIDC(I,96)=0
102 TRIDC(I,97)=0
103 TRIDC(I,98)=0
104 TRIDC(I,99)=0
105 TRIDC(I,100)=0

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155 15 JMILO,MSJDI
156 16 JMILO,MSJDI
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245 105 JMILO,MSJDI
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247 107 JMILO,MSJDI
248 108 JMILO,MSJDI
249 109 JMILO,MSJDI
250 110 JMILO,MSJDI


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390 741 50=9GAM(LP11)+GAM(LP11)OAREA/XDIF(I5)
391 811 50=9GAM(LP11)+GAM(LP11)OAREA/XDIF(I5)
392 811 50=9GAM(LP11)+GAM(LP11)OAREA/XDIF(I5)
393 IF (I5.EQ.2) DIST=XDIF(I2)
394 ALX=50(ALXOALX)
395 IF (I5.EQ.1) ALX=50(ALXOALX)
396 CUP=ALX
397 CUP=ALX
398 CUP=ALX
399 CUP=ALX
400 IF (I5.EQ.1) (I5.NE.1)+DFAC(I5)=I
401 L=I
402 L=I
403 L=I
404 L=I
405 L=I
406 L=I
407 L=I
408 L=I
409 L=I
410 L=I
411 L=I
412 L=I
413 L=I
414 L=I
415 L=I
416 L=I
417 L=I
418 L=I
419 L=I
420 IF (I5.EQ.1) GO TO 167
421 ALX=50(ALXOALX)+MO(LP11)OAREA
422 CUP=ALX
423 IF (I5.EQ.1) DIST=XDIF(I5)
424 IF (I5.EQ.1) DIST=XDIF(I5)
425 IF (I5.EQ.1) DIST=XDIF(I5)
426 IF (I5.EQ.1) DIST=XDIF(I5)
427 IF (I5.EQ.1) DIST=XDIF(I5)
428 IF (I5.EQ.1) DIST=XDIF(I5)
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458 IF (I5.EQ.1) DIST=XDIF(I5)
459 IF (I5.EQ.1) DIST=XDIF(I5)
460 IF (I5.EQ.1) DIST=XDIF(I5)

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455      IF (L1 .GT. 47)
456          GO TO 47
457          CALL IZP100(LZP100, LZP100AREA)
458          CALL IZP100(LZP100, LZP100AREA)
459          CALL IZP100(LZP100, LZP100AREA)
460          CALL IZP100(LZP100, LZP100AREA)
461          CALL IZP100(LZP100, LZP100AREA)
462          CALL IZP100(LZP100, LZP100AREA)
463          CALL IZP100(LZP100, LZP100AREA)
464          CALL IZP100(LZP100, LZP100AREA)
465          CALL IZP100(LZP100, LZP100AREA)
466          CALL IZP100(LZP100, LZP100AREA)
467          CALL IZP100(LZP100, LZP100AREA)
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471          CALL IZP100(LZP100, LZP100AREA)
472          CALL IZP100(LZP100, LZP100AREA)
473          CALL IZP100(LZP100, LZP100AREA)
474          CALL IZP100(LZP100, LZP100AREA)
475          CALL IZP100(LZP100, LZP100AREA)
476          CALL IZP100(LZP100, LZP100AREA)
477          CALL IZP100(LZP100, LZP100AREA)
478          CALL IZP100(LZP100, LZP100AREA)
479          CALL IZP100(LZP100, LZP100AREA)
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530          CALL IZP100(LZP100, LZP100AREA)
531          CALL IZP100(LZP100, LZP100AREA)
532          CALL IZP100(LZP100, LZP100AREA)
533          CALL IZP100(LZP100, LZP100AREA)

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695  CTR=NRMA(QULLEP)
696  CTR=NRMA(QULLEP)
697  DO 62 I=1,5
698  MAX=NRMA(QULLEP)
699  LPA=NRMA(QULLEP)
700  LPA=NRMA(QULLEP)
701  LPA=NRMA(QULLEP)
702  LPA=NRMA(QULLEP)
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798  LPA=NRMA(QULLEP)
799  LPA=NRMA(QULLEP)
800  LPA=NRMA(QULLEP)

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775 DO 81 J=1,NM1
776 L3=1+J*(NMP1)
777 L2=1+J*(NMP1)
778 CALL DSU(1,P2)
779 C 561 CUBIN01P2
780 C 561E (N1Z1)U*E*P F=VARIABLES
781 IF (N1-ME(LN1)) GO TO 710
782 REMA=711
783 C 561E P IS MC P IS DISSIPATION
784 C 710 IF (N1-ME(LN1)) GO TO 714
785 C 561E P IS MC P IS DISS
786 C 561E P IS MC P IS MC DU IS MC
787 C 714 IF (N1-ME(LN1)) GO TO 720
788 C 561E P IS MC P IS FAV
789 C 717 1=1
790 IF (N1-ME(LN1)) GO TO 712
791 L3=1+J*(NMP1)
792 L2=1+J*(NMP1)
793 L1=1+J*(NMP1)
794 L=1+J*(NMP1)
795 C 712 L1=L2=L3=L
796 L=1+J*(NMP1)
797 L=1+J*(NMP1)
798 L=1+J*(NMP1)
799 L=1+J*(NMP1)
800 C 717 1=1
801 L=1+J*(NMP1)
802 L=1+J*(NMP1)
803 L=1+J*(NMP1)
804 L=1+J*(NMP1)
805 C 717 1=1
806 L=1+J*(NMP1)
807 L=1+J*(NMP1)
808 L=1+J*(NMP1)
809 L=1+J*(NMP1)
810 C 760 L=1+J*(NMP1)
811 L=1+J*(NMP1)
812 L=1+J*(NMP1)
813 L=1+J*(NMP1)
814 L=1+J*(NMP1)
815 C 710 CONTINUE
816 IF (L3-VE(LN1)) GO TO 702,703
817 C 760 L=1+J*(NMP1)
818 C 760 L=1+J*(NMP1)
819 C 760 L=1+J*(NMP1)
820 C 760 L=1+J*(NMP1)
821 C 760 L=1+J*(NMP1)
822 C 760 L=1+J*(NMP1)
823 C 760 L=1+J*(NMP1)
824 C 760 L=1+J*(NMP1)
825 C 760 L=1+J*(NMP1)
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841 C 760 L=1+J*(NMP1)
842 C 760 L=1+J*(NMP1)
843 C 760 L=1+J*(NMP1)
844 C 760 L=1+J*(NMP1)
845 C 760 L=1+J*(NMP1)

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355 SUBROUTINE STRAD 78774 OPT=9 TRACE
CALL OPTIMIZE(LZP1)←B1.1
CONTINUE
END

PTN 4-6-439

08/24/78 14.08.46

PAGE 3

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101
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104

CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM

35 1
37 1
38 1

35 IS STATEMENT MAY BE MORE
COMMON OR EQUIVALENT THAN
37 IS STATEMENT MAY BE MORE
COMMON OR EQUIVALENT THAN
38 IS STATEMENT MAY BE MORE
COMMON OR EQUIVALENT THAN

OPTIMIZATION MAY BE
IMPROVED. OPTIMIZATION MAY BE
IMPROVED. OPTIMIZATION MAY BE
IMPROVED. OPTIMIZATION MAY BE
IMPROVED.

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1 SUBROUTINE SOLVE
COMMON/LS/DO(200),DF(200),D(200),R(200),C(200),APP(30),RPP(30)
1 DO=0,DF=0,D=0,R=0,C=0,APP=0,RPP=0
5 1 AX=1001,AY=1001,AXI=1001,AYI=1001,AXII=1001,AYII=1001,AXIII=1001,AYIII=1001
2 AX=1001,AY=1001,AXI=1001,AYI=1001,AXII=1001,AYII=1001,AXIII=1001,AYIII=1001
3 CZ=1001,CT=1001,CY=1001,CTI=1001,CYI=1001,CZII=1001,CYII=1001,CZIII=1001,CYIII=1001
4 CT=1001,CY=1001,CTI=1001,CYI=1001,CZII=1001,CYII=1001,CZIII=1001,CYIII=1001
10 1 DEGREE=20,ORDER=20,ITER=20,PRINT=0,DEBUG=0,TEMP=5001
EQUIVALENCE (DF,DF), (D,D), (R,R), (C,C), (APP,APP), (RPP,RPP)
COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
15 1 DO=0,DF=0,D=0,R=0,C=0,APP=0,RPP=0
2 F2=0,F3=0,F4=0,F5=0,F6=0,F7=0,F8=0,F9=0,F10=0,F11=0,F12=0,F13=0,F14=0,F15=0
3 COMMON/INTL/INTL(20),INTLI(20),INTLII(20),INTLIII(20),INTLIIII(20),INTLIIII(20)
20 1 SOLVE=0,ITER=0,ORDER=0,PRINT=0,DEBUG=0,TEMP=5001
3 AX=1001,AY=1001,AXI=1001,AYI=1001,AXII=1001,AYII=1001,AXIII=1001,AYIII=1001
4 AX=1001,AY=1001,AXI=1001,AYI=1001,AXII=1001,AYII=1001,AXIII=1001,AYIII=1001
5 AX=1001,AY=1001,AXI=1001,AYI=1001,AXII=1001,AYII=1001,AXIII=1001,AYIII=1001
25 1 PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1
2 PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1
3 PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1,PLANKI=1
30 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
35 1 DO=0,DF=0,D=0,R=0,C=0,APP=0,RPP=0
40 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
45 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
50 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
55 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
60 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
65 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
70 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)
75 1 COMMON/CT/CT(20),CY(20),CTI(20),CYI(20),CTII(20),CYII(20),CTIII(20),CYIII(20)
2 COMMON/CR/CR(20),CRI(20),CRII(20),CRII(20),CRIII(20),CRIIII(20)

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1  SUBROUTINE PRINT (ISTOP, ISTOP, ISTOP, ISTOP)
2  ISTOP = ISTOP + 1
3  ISTOP = ISTOP + 1
4  ISTOP = ISTOP + 1
5  ISTOP = ISTOP + 1
6  ISTOP = ISTOP + 1
7  ISTOP = ISTOP + 1
8  ISTOP = ISTOP + 1
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71 ISTOP = ISTOP + 1
72 ISTOP = ISTOP + 1

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STOP NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM CONTROL VARIABLE IN COMMON ON EQUILIBRENCED. OPTIMIZATION MAY BE INTERRUPTED.

SUBROUTINE FPRINT 7376 OPT=0 TRACE FTN 1.00439 09/26/79 14.08.66 PAGE 2
CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM
41 CONTROL VARIABLE IN COMMON OR EQUIVALENCED, OPTIMIZATION MAY BE INTERRUPTED.

APPENDIX D

LISTING OF LINER COOLING MODEL

PAGE 1

08/24/78 13.44.16

SYM 4-4439

73.74 OPT=0 TRACE

```

1  PROGRAM MAIN (INPUT,OUTPUT,TAPES,INPUT,TAPES,OUTPUT)
4  *****
5  *****
6  *****
7  *****
8  *****
9  *****
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99 *****
100 *****

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```

235 READ (5,4) (PFC(I),I=1,NUM)
240 READ (5,4) (FCO(I),I=1,NUM)
245 READ (5,4) (C(I),I=1,NUM)
250 READ (5,4) (S(I),I=1,NUM)
255 READ (5,4) (P(I),I=1,NUM)
260 READ (5,4) (L(I),I=1,NUM)
265 READ (5,4) (R(I),I=1,NUM)
270 READ (5,4) (T(I),I=1,NUM)
275 READ (5,4) (X(I),I=1,NUM)
280 READ (5,4) (Y(I),I=1,NUM)
285 READ (5,4) (Z(I),I=1,NUM)
290 READ (5,4) (A(I),I=1,NUM)
295 READ (5,4) (B(I),I=1,NUM)
300 READ (5,4) (C(I),I=1,NUM)
305 READ (5,4) (D(I),I=1,NUM)

```


FTN 4.0-43C

73774 CRT=C TRAC

PROGRAM MAIN 73774 CRT=C TRAC

1-10-78 10:10:10

08/26/78 13.44.16

7

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665      1-10-78 10:10:10
670      1-10-78 10:10:10
675      1-10-78 10:10:10
680      1-10-78 10:10:10
685      1-10-78 10:10:10
690      1-10-78 10:10:10
695      1-10-78 10:10:10
700      1-10-78 10:10:10
705      1-10-78 10:10:10
710      1-10-78 10:10:10
715      1-10-78 10:10:10
720      1-10-78 10:10:10
725      1-10-78 10:10:10
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735      1-10-78 10:10:10
740      1-10-78 10:10:10
745      1-10-78 10:10:10
750      1-10-78 10:10:10
755      1-10-78 10:10:10
760      1-10-78 10:10:10
765      1-10-78 10:10:10
770      1-10-78 10:10:10
775      1-10-78 10:10:10
780      1-10-78 10:10:10
785      1-10-78 10:10:10
790      1-10-78 10:10:10
795      1-10-78 10:10:10
800      1-10-78 10:10:10
805      1-10-78 10:10:10
810      1-10-78 10:10:10
815      1-10-78 10:10:10
820      1-10-78 10:10:10
825      1-10-78 10:10:10
830      1-10-78 10:10:10
835      1-10-78 10:10:10
840      1-10-78 10:10:10
845      1-10-78 10:10:10
850      1-10-78 10:10:10
855      1-10-78 10:10:10
860      1-10-78 10:10:10
865      1-10-78 10:10:10
870      1-10-78 10:10:10
875      1-10-78 10:10:10
880      1-10-78 10:10:10
885      1-10-78 10:10:10
890      1-10-78 10:10:10
895      1-10-78 10:10:10
900      1-10-78 10:10:10
905      1-10-78 10:10:10
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915      1-10-78 10:10:10
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930      1-10-78 10:10:10
935      1-10-78 10:10:10
940      1-10-78 10:10:10
945      1-10-78 10:10:10
950      1-10-78 10:10:10
955      1-10-78 10:10:10
960      1-10-78 10:10:10
965      1-10-78 10:10:10
970      1-10-78 10:10:10
975      1-10-78 10:10:10
980      1-10-78 10:10:10
985      1-10-78 10:10:10
990      1-10-78 10:10:10
995      1-10-78 10:10:10

```

----- FURTHER ADJUSTMENTS TO BE MADE MADE IN CHAPTERS P AND Q.

546

PA

525

CHAPTER-----

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586

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620 620 VEZ=AVM(J(MV1+2,VUMJ(MV1+2)))
      621 IF (AVM(J(MV1+2,VUMJ(MV1+2)))>.5)ALM
      622 F1(I,1)=VUMJ(MV1+2)
      623 MDO(I)=MD(VUMJ(MV1+2),1)
      624 RUL(I)=RMD(VUMJ(MV1+2),1)
      625 F1(I)=VUMJ(MV1+2)+.5*(AVM(J(MV1+2,VUMJ(MV1+2)))
      626 CONTINUE
      627 C 392
      628 IF (MD=LE.VUMJ(MV1)) GO TO 394
      629 IF (ABS(AVUMJ(MV1)-SUM3)-GT.1.E-10) GO TO 390
      630 MVI=MV1+1
      631 GO TO 394
      632 IF (SUM3=Z) GO TO 392
      633 SMI=AVUMJ(MV1+2)
      634 SMI=AVUMJ(MV1)
      635 MVI=AVUMJ(MV1+2)
      636 Y=VUMJ(MV1)
      637 F1(I)=AVUMJ(MV1)
      638 F1(I)=AVUMJ(MV1)
      639 DO 200 JJ=1,MSPM
      640 F1(I)=0
      641 F1(I)=0
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ROUTED
APUD= (EMP21002-P11302005
I= (RAD=EC.01 ADUB- (EMP31-V111
775 87 81M= 50011002
78 7E (1STEP- (OUT) ADUB- 50001002
780 81U= (EMP31002-2- (IN
I= (RAD=EC.01 AFLU- (EMP31-V111
81P= (AFLU- (EMP31-V111
81R= (AFLU- (EMP31-V111
790 81S= (STEP- (E) 50 TO 00
I= (STEP- (E) 50 TO 00
82 81T= (STEP- (E) 50 TO 00
83 81U= (STEP- (E) 50 TO 00
84 81V= (STEP- (E) 50 TO 00
85 81W= (STEP- (E) 50 TO 00
86 81X= (STEP- (E) 50 TO 00
87 81Y= (STEP- (E) 50 TO 00
88 81Z= (STEP- (E) 50 TO 00
89 820= (STEP- (E) 50 TO 00
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268 999= (STEP- (E) 50 TO 00
269 1000= (STEP- (E) 50 TO 00

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925 925 IF (IHEX(WE-ZI) GO TO 97
    926 926 GAT=ABS(IUP11-UIMP33)/(COMP31-(I1)*1.E-30))
    927 927 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    928 928 IF (I102) GO TO 99
    929 929 I1=I1+I1
    930 930 IF (IABS(WRT)+ABS(COMP1)+ABS(WPE) GO TO 96
    931 931 D=ABS(WPE)/(I1+ABS(WRT)+ABS(COMP1))
    932 932 CONTINUE
    933 933 IF (I102) GO TO 91
    934 934 IF (IABS(WRT)+ABS(COMP1)+ABS(WPE) GO TO 91
    935 935 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    936 936 IF (I102) GO TO 92
    937 937 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    938 938 CONTINUE
    939 939 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    940 940 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    941 941 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    942 942 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
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    967 967 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
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    970 970 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
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    999 999 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
    1000 1000 WRT=UM1+UM2+UM3+UM4+UM5+UM6+UM7+UM8+UM9+UM10
  
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1080	C	FLUX(JU)=FLUX(JU)-P*IE*E20+52A05TI FLUX(JM)=FLUX(JM)-P*IE*E1D+51A95TI	MA	1085
1081		WRITE (6,1031) X=FLUX(JU),FLUX(JM),P*IE	MA	1086
1082	C	FORMAT ('2E13.1, /E11.3,5E9.4 FLUX(J)') 1083 L (6E11.3)	MA	1087
1084		IF (X-0.0) GO TO 1024	MA	1088
1085		DD 1025 J=JUM*PUMEF	MA	1089
1086		IF (JUM*VEJ)-E0.0) GO TO 1025	MA	1090
1087		IF ((JUM*VEJ)-E0.0) GO TO 1025	MA	1091
1088		ADJ(1)=AJC(J)/BUREF(RF(J))	MA	1092
1089		ADJ(2)=AJC(J)/BUREF(RF(J))	MA	1093
1090		ADJ(3)=AJC(J)/BUREF(RF(J))	MA	1094
1091		ADJ(4)=AJC(J)/BUREF(RF(J))	MA	1095
1092		ADJ(5)=AJC(J)/BUREF(RF(J))	MA	1096
1093		ADJ(6)=AJC(J)/BUREF(RF(J))	MA	1097
1094		ADJ(7)=AJC(J)/BUREF(RF(J))	MA	1098
1095		ADJ(8)=AJC(J)/BUREF(RF(J))	MA	1099
1096		ADJ(9)=AJC(J)/BUREF(RF(J))	MA	1100
1097		ADJ(10)=AJC(J)/BUREF(RF(J))	MA	1101
1098		ADJ(11)=AJC(J)/BUREF(RF(J))	MA	1102
1099		ADJ(12)=AJC(J)/BUREF(RF(J))	MA	1103
1100		ADJ(13)=AJC(J)/BUREF(RF(J))	MA	1104
1101		ADJ(14)=AJC(J)/BUREF(RF(J))	MA	1105
1102		ADJ(15)=AJC(J)/BUREF(RF(J))	MA	1106
1103		ADJ(16)=AJC(J)/BUREF(RF(J))	MA	1107
1104		ADJ(17)=AJC(J)/BUREF(RF(J))	MA	1108
1105		ADJ(18)=AJC(J)/BUREF(RF(J))	MA	1109
1106		ADJ(19)=AJC(J)/BUREF(RF(J))	MA	1110
1107		ADJ(20)=AJC(J)/BUREF(RF(J))	MA	1111
1108		ADJ(21)=AJC(J)/BUREF(RF(J))	MA	1112
1109		ADJ(22)=AJC(J)/BUREF(RF(J))	MA	1113
1110		ADJ(23)=AJC(J)/BUREF(RF(J))	MA	1114
1111		ADJ(24)=AJC(J)/BUREF(RF(J))	MA	1115
1112		ADJ(25)=AJC(J)/BUREF(RF(J))	MA	1116
1113		ADJ(26)=AJC(J)/BUREF(RF(J))	MA	1117
1114		ADJ(27)=AJC(J)/BUREF(RF(J))	MA	1118
1115		ADJ(28)=AJC(J)/BUREF(RF(J))	MA	1119
1116		ADJ(29)=AJC(J)/BUREF(RF(J))	MA	1120
1117		ADJ(30)=AJC(J)/BUREF(RF(J))	MA	1121
1118		ADJ(31)=AJC(J)/BUREF(RF(J))	MA	1122
1119		ADJ(32)=AJC(J)/BUREF(RF(J))	MA	1123
1120		ADJ(33)=AJC(J)/BUREF(RF(J))	MA	1124
1121		ADJ(34)=AJC(J)/BUREF(RF(J))	MA	1125
1122		ADJ(35)=AJC(J)/BUREF(RF(J))	MA	1126
1123		ADJ(36)=AJC(J)/BUREF(RF(J))	MA	1127
1124		ADJ(37)=AJC(J)/BUREF(RF(J))	MA	1128
1125		ADJ(38)=AJC(J)/BUREF(RF(J))	MA	1129
1126		ADJ(39)=AJC(J)/BUREF(RF(J))	MA	1130
1127		ADJ(40)=AJC(J)/BUREF(RF(J))	MA	1131
1128		ADJ(41)=AJC(J)/BUREF(RF(J))	MA	1132
1129		ADJ(42)=AJC(J)/BUREF(RF(J))	MA	1133
1130		ADJ(43)=AJC(J)/BUREF(RF(J))	MA	1134
1131		ADJ(44)=AJC(J)/BUREF(RF(J))	MA	1135
1132		ADJ(45)=AJC(J)/BUREF(RF(J))	MA	1136
1133		ADJ(46)=AJC(J)/BUREF(RF(J))	MA	1137
1134		ADJ(47)=AJC(J)/BUREF(RF(J))	MA	1138
1135		ADJ(48)=AJC(J)/BUREF(RF(J))	MA	1139
1136		ADJ(49)=AJC(J)/BUREF(RF(J))	MA	1140
1137		ADJ(50)=AJC(J)/BUREF(RF(J))	MA	1141
1138		ADJ(51)=AJC(J)/BUREF(RF(J))	MA	1142
1139		ADJ(52)=AJC(J)/BUREF(RF(J))	MA	1143
1140		ADJ(53)=AJC(J)/BUREF(RF(J))	MA	1144
1141		ADJ(54)=AJC(J)/BUREF(RF(J))	MA	1145
1142		ADJ(55)=AJC(J)/BUREF(RF(J))	MA	1146
1143		ADJ(56)=AJC(J)/BUREF(RF(J))	MA	1147
1144		ADJ(57)=AJC(J)/BUREF(RF(J))	MA	1148
1145		ADJ(58)=AJC(J)/BUREF(RF(J))	MA	1149
1146		ADJ(59)=AJC(J)/BUREF(RF(J))	MA	1150
1147		ADJ(60)=AJC(J)/BUREF(RF(J))	MA	1151
1148		ADJ(61)=AJC(J)/BUREF(RF(J))	MA	1152
1149		ADJ(62)=AJC(J)/BUREF(RF(J))	MA	1153
1150		ADJ(63)=AJC(J)/BUREF(RF(J))	MA	1154
1151		ADJ(64)=AJC(J)/BUREF(RF(J))	MA	1155

0072470 13.44.14

RTM 4.4430

FUNCTION WISCO 7374 2010C 1025F

1A	25
1A	26
1A	27
1A	28
1A	29
1A	30
1A	31
1A	32

```

1  FUNCTION WISCO(7)
   IF (1.5:2007.1 60 10 10
   WISCO=1.808E-5(17204.61007.74
   5  10 WISCO=4.444E-5(55507171111.1
      RETURN
   END

```

```

1  SUBROUTINE A05 (N, X, Y, Z, W, V, U, T, S, R, Q, P, O, N, M, L, K, J, I, H, G, F, E, D, C, B, A)
2  DIMENSION X(100), Y(100), Z(100), W(100), V(100), U(100), T(100), S(100), R(100), Q(100), P(100), O(100), N(100), M(100), L(100), K(100), J(100), I(100), H(100), G(100), F(100), E(100), D(100), C(100), B(100), A(100)
3  COMMON /A05/ X, Y, Z, W, V, U, T, S, R, Q, P, O, N, M, L, K, J, I, H, G, F, E, D, C, B, A
4  IF (N.EQ.0) GO TO 10
5  DO 10 I=1,N
6  X(I)=X(I)+1
7  Y(I)=Y(I)+1
8  Z(I)=Z(I)+1
9  W(I)=W(I)+1
10 V(I)=V(I)+1
11 U(I)=U(I)+1
12 T(I)=T(I)+1
13 S(I)=S(I)+1
14 R(I)=R(I)+1
15 Q(I)=Q(I)+1
16 P(I)=P(I)+1
17 O(I)=O(I)+1
18 N(I)=N(I)+1
19 M(I)=M(I)+1
20 L(I)=L(I)+1
21 K(I)=K(I)+1
22 J(I)=J(I)+1
23 I(I)=I(I)+1
24 H(I)=H(I)+1
25 G(I)=G(I)+1
26 F(I)=F(I)+1
27 E(I)=E(I)+1
28 D(I)=D(I)+1
29 C(I)=C(I)+1
30 B(I)=B(I)+1
31 A(I)=A(I)+1
32 GO TO 10
33 END

```

```

90      Y=11-1//PDEF(1180.5
91      SUII=55 DIII=5 EII=11307
92      20 33 102.791
93      DO 91C GO TO 941
94      DO 91D GO TO 942
95      FLUTE=516785(11,1104
96      SOR=2000500(11,1104
97      TD=151700(11,1104)
98      COW=500(11,1104)
99      SUII=55 DIII=5 EII=11307
100     SUII=55 DIII=5 EII=11307
101     SUII=55 DIII=5 EII=11307
102     SUII=55 DIII=5 EII=11307
103     SUII=55 DIII=5 EII=11307
104     SUII=55 DIII=5 EII=11307
105     SUII=55 DIII=5 EII=11307
106     SUII=55 DIII=5 EII=11307
107     SUII=55 DIII=5 EII=11307
108     SUII=55 DIII=5 EII=11307
109     SUII=55 DIII=5 EII=11307
110     SUII=55 DIII=5 EII=11307
111     SUII=55 DIII=5 EII=11307
112     SUII=55 DIII=5 EII=11307
113     SUII=55 DIII=5 EII=11307
114     SUII=55 DIII=5 EII=11307
115     SUII=55 DIII=5 EII=11307
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117     SUII=55 DIII=5 EII=11307
118     SUII=55 DIII=5 EII=11307
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141     SUII=55 DIII=5 EII=11307
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148     SUII=55 DIII=5 EII=11307
149     SUII=55 DIII=5 EII=11307
150     SUII=55 DIII=5 EII=11307
151     SUII=55 DIII=5 EII=11307
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153     SUII=55 DIII=5 EII=11307
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179     SUII=55 DIII=5 EII=11307
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181     SUII=55 DIII=5 EII=11307
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183     SUII=55 DIII=5 EII=11307
184     SUII=55 DIII=5 EII=11307
185     SUII=55 DIII=5 EII=11307
186     SUII=55 DIII=5 EII=11307
187     SUII=55 DIII=5 EII=11307
188     SUII=55 DIII=5 EII=11307
189     SUII=55 DIII=5 EII=11307
190     SUII=55 DIII=5 EII=11307
191     SUII=55 DIII=5 EII=11307
192     SUII=55 DIII=5 EII=11307
193     SUII=55 DIII=5 EII=11307
194     SUII=55 DIII=5 EII=11307
195     SUII=55 DIII=5 EII=11307
196     SUII=55 DIII=5 EII=11307
197     SUII=55 DIII=5 EII=11307
198     SUII=55 DIII=5 EII=11307
199     SUII=55 DIII=5 EII=11307
200     SUII=55 DIII=5 EII=11307

```

08/26/78 13-44-16

878 6-6-639

7377* OPT=0 TRACE

Address	Subroutine	Instruction	Hex	Hex	Hex	Hex
155	SUBROUTINE SUB	SWT(1)=0	40	10	40	
160		SWT(1)=0	40	10	40	
165		SWT(1)=0	40	10	40	
170		SWT(1)=0	40	10	40	
175		SWT(1)=0	40	10	40	
180		SWT(1)=0	40	10	40	
185		SWT(1)=0	40	10	40	
190		SWT(1)=0	40	10	40	
195		SWT(1)=0	40	10	40	
200		SWT(1)=0	40	10	40	
205		SWT(1)=0	40	10	40	
210		SWT(1)=0	40	10	40	
215		SWT(1)=0	40	10	40	
220		SWT(1)=0	40	10	40	
225		SWT(1)=0	40	10	40	
230		SWT(1)=0	40	10	40	

LINE NO	ADDRESS	DATA
310	00000000	00000000
315	00000000	00000000
320	00000000	00000000
325	00000000	00000000
330	00000000	00000000
335	00000000	00000000
340	00000000	00000000
345	00000000	00000000
350	00000000	00000000
355	00000000	00000000
360	00000000	00000000
365	00000000	00000000
370	00000000	00000000
375	00000000	00000000
380	00000000	00000000

04/24/79 17.44.16

STN 4.2+438

SUBROUTINE AUX 73/74 COT=0 TRACE

390 LAB(2)=-AM
 WRITE (6,100) LAB(CSDEI),I=1,MP3)
 LAB(2)=-MSDEL
 LAB(2)=-MI
 LAB(2)=-1001 LAB(CSDEI),I=1,MP3)
 CONTINUE
 89
 C 6000 CONTINUE
 RETURN
 100 END
 395 PERMILLIM *A5*42*101E11-3/798*11F11-3))

372 AU
 373 AU
 374 AU
 375 AU
 376 AU
 377 AU
 378 AU
 379 AU
 380 AU
 381 AU
 382 AU

CARD NO.	SEVERITY	DETAILS	DIAGNOSIS OF PROBLEM
141	1	LAB(2)=-AM	AN IF STATEMENT MAY BE
142	1	WRITE (6,100) LAB(CSDEI),I=1,MP3)	THE PRESENT MAY BE
143	1	LAB(2)=-MSDEL	THE PRESENT MAY BE
144	1	LAB(2)=-MI	THE PRESENT MAY BE
145	1	LAB(2)=-1001 LAB(CSDEI),I=1,MP3)	THE PRESENT MAY BE
146	1	CONTINUE	THE PRESENT MAY BE
147	1	89	THE PRESENT MAY BE
148	1	C 6000 CONTINUE	THE PRESENT MAY BE
149	1	RETURN	THE PRESENT MAY BE
150	1	100 END	THE PRESENT MAY BE
151	1	395 PERMILLIM *A5*42*101E11-3/798*11F11-3))	THE PRESENT MAY BE

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1 1340V THE 906
2 1341V 0000 1342V 1343V 1344V 1345V 1346V 1347V 1348V 1349V 1350V
3 1351V 1352V 1353V 1354V 1355V 1356V 1357V 1358V 1359V 1360V
4 1361V 1362V 1363V 1364V 1365V 1366V 1367V 1368V 1369V 1370V
5 1371V 1372V 1373V 1374V 1375V 1376V 1377V 1378V 1379V 1380V
6 1381V 1382V 1383V 1384V 1385V 1386V 1387V 1388V 1389V 1390V
7 1391V 1392V 1393V 1394V 1395V 1396V 1397V 1398V 1399V 1400V
8 1401V 1402V 1403V 1404V 1405V 1406V 1407V 1408V 1409V 1410V
9 1411V 1412V 1413V 1414V 1415V 1416V 1417V 1418V 1419V 1420V
10 1421V 1422V 1423V 1424V 1425V 1426V 1427V 1428V 1429V 1430V
11 1431V 1432V 1433V 1434V 1435V 1436V 1437V 1438V 1439V 1440V
12 1441V 1442V 1443V 1444V 1445V 1446V 1447V 1448V 1449V 1450V
13 1451V 1452V 1453V 1454V 1455V 1456V 1457V 1458V 1459V 1460V
14 1461V 1462V 1463V 1464V 1465V 1466V 1467V 1468V 1469V 1470V
15 1471V 1472V 1473V 1474V 1475V 1476V 1477V 1478V 1479V 1480V
16 1481V 1482V 1483V 1484V 1485V 1486V 1487V 1488V 1489V 1490V
17 1491V 1492V 1493V 1494V 1495V 1496V 1497V 1498V 1499V 1500V
18 1501V 1502V 1503V 1504V 1505V 1506V 1507V 1508V 1509V 1510V
19 1511V 1512V 1513V 1514V 1515V 1516V 1517V 1518V 1519V 1520V
20 1521V 1522V 1523V 1524V 1525V 1526V 1527V 1528V 1529V 1530V
21 1531V 1532V 1533V 1534V 1535V 1536V 1537V 1538V 1539V 1540V
22 1541V 1542V 1543V 1544V 1545V 1546V 1547V 1548V 1549V 1550V
23 1551V 1552V 1553V 1554V 1555V 1556V 1557V 1558V 1559V 1560V
24 1561V 1562V 1563V 1564V 1565V 1566V 1567V 1568V 1569V 1570V
25 1571V 1572V 1573V 1574V 1575V 1576V 1577V 1578V 1579V 1580V
26 1581V 1582V 1583V 1584V 1585V 1586V 1587V 1588V 1589V 1590V
27 1591V 1592V 1593V 1594V 1595V 1596V 1597V 1598V 1599V 1600V
28 1601V 1602V 1603V 1604V 1605V 1606V 1607V 1608V 1609V 1610V
29 1611V 1612V 1613V 1614V 1615V 1616V 1617V 1618V 1619V 1620V
30 1621V 1622V 1623V 1624V 1625V 1626V 1627V 1628V 1629V 1630V
31 1631V 1632V 1633V 1634V 1635V 1636V 1637V 1638V 1639V 1640V
32 1641V 1642V 1643V 1644V 1645V 1646V 1647V 1648V 1649V 1650V
33 1651V 1652V 1653V 1654V 1655V 1656V 1657V 1658V 1659V 1660V
34 1661V 1662V 1663V 1664V 1665V 1666V 1667V 1668V 1669V 1670V
35 1671V 1672V 1673V 1674V 1675V 1676V 1677V 1678V 1679V 1680V
36 1681V 1682V 1683V 1684V 1685V 1686V 1687V 1688V 1689V 1690V
37 1691V 1692V 1693V 1694V 1695V 1696V 1697V 1698V 1699V 1700V
38 1701V 1702V 1703V 1704V 1705V 1706V 1707V 1708V 1709V 1710V
39 1711V 1712V 1713V 1714V 1715V 1716V 1717V 1718V 1719V 1720V
40 1721V 1722V 1723V 1724V 1725V 1726V 1727V 1728V 1729V 1730V
41 1731V 1732V 1733V 1734V 1735V 1736V 1737V 1738V 1739V 1740V
42 1741V 1742V 1743V 1744V 1745V 1746V 1747V 1748V 1749V 1750V
43 1751V 1752V 1753V 1754V 1755V 1756V 1757V 1758V 1759V 1760V
44 1761V 1762V 1763V 1764V 1765V 1766V 1767V 1768V 1769V 1770V
45 1771V 1772V 1773V 1774V 1775V 1776V 1777V 1778V 1779V 1780V
46 1781V 1782V 1783V 1784V 1785V 1786V 1787V 1788V 1789V 1790V
47 1791V 1792V 1793V 1794V 1795V 1796V 1797V 1798V 1799V 1800V
48 1801V 1802V 1803V 1804V 1805V 1806V 1807V 1808V 1809V 1810V
49 1811V 1812V 1813V 1814V 1815V 1816V 1817V 1818V 1819V 1820V
50 1821V 1822V 1823V 1824V 1825V 1826V 1827V 1828V 1829V 1830V
51 1831V 1832V 1833V 1834V 1835V 1836V 1837V 1838V 1839V 1840V
52 1841V 1842V 1843V 1844V 1845V 1846V 1847V 1848V 1849V 1850V
53 1851V 1852V 1853V 1854V 1855V 1856V 1857V 1858V 1859V 1860V
54 1861V 1862V 1863V 1864V 1865V 1866V 1867V 1868V 1869V 1870V
55 1871V 1872V 1873V 1874V 1875V 1876V 1877V 1878V 1879V 1880V
56 1881V 1882V 1883V 1884V 1885V 1886V 1887V 1888V 1889V 1890V
57 1891V 1892V 1893V 1894V 1895V 1896V 1897V 1898V 1899V 1900V
58 1901V 1902V 1903V 1904V 1905V 1906V 1907V 1908V 1909V 1910V
59 1911V 1912V 1913V 1914V 1915V 1916V 1917V 1918V 1919V 1920V
60 1921V 1922V 1923V 1924V 1925V 1926V 1927V 1928V 1929V 1930V
61 1931V 1932V 1933V 1934V 1935V 1936V 1937V 1938V 1939V 1940V
62 1941V 1942V 1943V 1944V 1945V 1946V 1947V 1948V 1949V 1950V
63 1951V 1952V 1953V 1954V 1955V 1956V 1957V 1958V 1959V 1960V
64 1961V 1962V 1963V 1964V 1965V 1966V 1967V 1968V 1969V 1970V
65 1971V 1972V 1973V 1974V 1975V 1976V 1977V 1978V 1979V 1980V
66 1981V 1982V 1983V 1984V 1985V 1986V 1987V 1988V 1989V 1990V
67 1991V 1992V 1993V 1994V 1995V 1996V 1997V 1998V 1999V 2000V
68 1999V 2000V
69 2000V
70 2000V
71 2000V
72 2000V
73 2000V
74 2000V
75 2000V

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```

155   DVE=DTM
    IROUND=-1
    GO TO 240
160   DVE=TA-DLCL(I)
    DVE=DVE-DIFD(I)*DPP/DTADCS*SMALL
    IROUND=1
165   DVE=DTM-DXFD+DVE*DTF3
    IF (ABS(DVE-DST)-ABS(-D1+DSC(I))) GO TO 180
170   DVE=DVE+DVE*DTF
    IF (IROUND) DVE=DVE*(1+VJ4-I)
    GO TO 310
175   DVE=DTM*DTF+DVE*(J+I-VJ4-I)
    DVE=DVE+DVE*DTF
    FEVAP(I)=1.0
    GO TO 320
180   L=LN(L*E)
    FEVAP(I)=FEVAP(I)+DVE*DTM
    FEVAP(I)=FEVAP(I)-L*DTM
    GO TO 320
185   FEVAP(I)=1.0
    L=LN(L*E)
    FEVAP(I)=FEVAP(I)+L*DTM
    GO TO 330
190   DVE=DTM
    FEVAP(I)=1.0
195   IF (I*DTM+.5) GO TO 340
    WRITE (99001) DEAP,FEVAP(I),DTM,DTM*DTM*DTM
200   IF (DVE+L+.5) GO TO 350
    FEVAP(I)=FEVAP(I)+DVE*DTM
    DVE=DTM
    JK=JK+1
    RETURN
    END

```

LINE	FUNCTION	DESCRIPTION	AMOUNT	DATE	TIME	STATUS	ENTRY
1	C	FUNCTION GASP (TTL)					
2	C	DIMENSION 4M(7,12),AL(7,12),R(12)					
3	C	SPECIES GROUP - FUEL,CO2,CO,72,420,04,7,4,42,8,40,42					
4	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
5	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
6	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
7	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
8	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
9	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
10	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
11	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
12	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
13	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
14	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
15	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
16	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
17	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
18	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
19	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
20	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
21	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
22	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
23	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
24	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
25	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
26	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
27	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
28	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
29	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
30	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
31	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
32	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
33	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
34	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
35	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
36	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
37	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
38	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
39	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
40	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
41	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
42	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
43	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
44	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
45	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
46	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
47	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
48	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
49	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
50	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
51	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
52	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
53	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
54	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
55	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
56	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
57	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
58	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
59	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
60	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
61	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
62	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
63	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
64	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
65	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
66	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
67	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
68	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
69	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
70	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
71	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
72	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
73	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
74	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
75	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
76	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
77	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					
78	C	DATA (NAME) J1, I=1, 71, 47, 12, 17					


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1 SUBROUTINE STB2D
2     DIMENSION STAB(40), STAC(40), STAD(40), STAE(40), STAF(40), STAG(40), STAH(40), STAI(40), STAJ(40), STAK(40),
3     STAL(40), STAM(40), STAN(40), STAO(40), STAP(40), STAQ(40), STAR(40), STAS(40), STAT(40), STAU(40), STAV(40),
4     STAW(40), STAX(40), STAY(40), STAZ(40), STBA(40), STBC(40), STBD(40), STBE(40), STBF(40), STBG(40), STBH(40),
5     STBI(40), STBJ(40), STBK(40), STBL(40), STBM(40), STBN(40), STBO(40), STBP(40), STBQ(40), STBR(40), STBS(40), STBT(40),
6     STBU(40), STBV(40), STBW(40), STBX(40), STBY(40), STBZ(40), STCA(40), STCB(40), STCC(40), STCE(40), STCF(40), STCG(40),
7     STCH(40), STCI(40), STCJ(40), STCK(40), STCL(40), STCM(40), STCN(40), STCO(40), STCP(40), STCQ(40), STCR(40), STCS(40),
8     STCT(40), STCU(40), STCV(40), STCW(40), STCX(40), STCY(40), STCZ(40), STDA(40), STDB(40), STDC(40), STDE(40), STDF(40),
9     STDG(40), STDH(40), STDI(40), STDJ(40), STDK(40), STDL(40), STDM(40), STDN(40), STDO(40), STDP(40), STDQ(40), STDR(40),
10    STDS(40), STDT(40), STDU(40), STDV(40), STDW(40), STDX(40), STDY(40), STDZ(40), STEA(40), STEB(40), STEC(40),
11    STEF(40), STEG(40), STEH(40), STEI(40), STEJ(40), STEK(40), STEL(40), STEM(40), STEN(40), STEO(40), STEP(40),
12    STEQ(40), STER(40), STES(40), STET(40), STEU(40), STEV(40), STEW(40), STEX(40), STEY(40), STEZ(40), STFA(40),
13    STFB(40), STFC(40), STFD(40), STFE(40), STFF(40), STFG(40), STFH(40), STFI(40), STFJ(40), STFK(40), STFL(40),
14    STFM(40), STFN(40), STFO(40), STFP(40), STFQ(40), STFR(40), STFS(40), STFT(40), STFU(40), STFV(40), STFW(40),
15    STFX(40), STFY(40), STFZ(40), STGA(40), STGB(40), STGC(40), STGD(40), STGE(40), STGF(40), STGH(40), STGI(40),
16    STGJ(40), STGK(40), STGL(40), STGM(40), STGN(40), STGO(40), STGP(40), STGQ(40), STGR(40), STGS(40), STGT(40),
17    STGU(40), STGV(40), STGW(40), STGX(40), STGY(40), STGZ(40), STHA(40), STHB(40), STHC(40), STHD(40), STEH(40),
18    STHF(40), STHG(40), SHI(40), SHJ(40), SHK(40), SHL(40), SHM(40), SHN(40), SHO(40), SHP(40), SHQ(40), SHR(40),
19    SHS(40), SHT(40), SHU(40), SHV(40), SHW(40), SHX(40), SHY(40), SHZ(40), STIA(40), STIB(40), STIC(40), STID(40),
20    STIE(40), STIF(40), STIG(40), STIH(40), STII(40), STIJ(40), STIK(40), STIL(40), STIM(40), STIN(40), STIO(40),
21    STIP(40), STIQ(40), STIR(40), STIS(40), STIT(40), STIU(40), STIV(40), STIW(40), STIX(40), STIY(40), STIZ(40),
22    STJA(40), STJB(40), STJC(40), STJD(40), STJE(40), STJF(40), STJG(40), STJH(40), STJI(40), STJJ(40), STJK(40),
23    STJL(40), STJM(40), STJN(40), STJO(40), STJP(40), STJQ(40), STJR(40), STJS(40), STJT(40), STJU(40), STJV(40),
24    STJW(40), STJX(40), STJY(40), STJZ(40), STKA(40), STKB(40), STKC(40), STKD(40), STEK(40), STKF(40), STKG(40),
25    STKH(40), STKI(40), STKJ(40), STKK(40), STKL(40), STKM(40), STKN(40), STKO(40), STKP(40), STKQ(40), STKR(40),
26    STKS(40), STKT(40), STKU(40), STKV(40), STKW(40), STKX(40), STKY(40), STKZ(40), STLA(40), STLB(40), STLC(40),
27    STLD(40), STEL(40), STLF(40), STLG(40), SLA(40), SLB(40), SLC(40), SLD(40), SLE(40), SLF(40), SLG(40), SLH(40),
28    SLI(40), SLJ(40), SLK(40), SLL(40), SLM(40), SLN(40), SLO(40), SLP(40), SLQ(40), SLR(40), SLS(40), SLT(40),
29    SLU(40), SLV(40), SLW(40), SLX(40), SLY(40), SLZ(40), STMA(40), STMB(40), STMC(40), STMD(40), STME(40), STMF(40),
30    STMG(40), STMH(40), STMI(40), STMJ(40), STMK(40), STML(40), STMM(40), STMN(40), STMO(40), STMP(40), STMQ(40),
31    STMR(40), STMS(40), STMT(40), STMU(40), STMV(40), STMW(40), STMX(40), STMY(40), STMZ(40), STNA(40), STNB(40),
32    STNC(40), STND(40), STNE(40), STNF(40), STNG(40), STNH(40), STNI(40), STNJ(40), STNK(40), STNL(40), STNM(40),
33    STNO(40), STNP(40), STNQ(40), STNR(40), STNS(40), STNT(40), STNU(40), STNV(40), STNW(40), STNX(40), STNY(40),
34    STNZ(40), STOA(40), STOB(40), STOC(40), STOD(40), STOE(40), STOF(40), STOG(40), STOH(40), STOI(40), STOJ(40),
35    STOK(40), STOL(40), STOM(40), STON(40), STOO(40), STOP(40), STOQ(40), STOR(40), STOS(40), STOT(40), STOU(40),
36    STOV(40), STOW(40), STOX(40), STOY(40), STOZ(40), STPA(40), STPB(40), STPC(40), STPD(40), STEP(40), STPF(40),
37    STPG(40), STPH(40), STPI(40), STPJ(40), STPK(40), STPL(40), STPM(40), STPN(40), STPO(40), STPP(40), STPQ(40),
38    STPR(40), STPS(40), STPT(40), STPU(40), STPV(40), STPW(40), STPX(40), STPY(40), STPZ(40), STQA(40), STQB(40),
39    STQC(40), STQD(40), STQE(40), STQF(40), STQG(40), STQH(40), STQI(40), STQJ(40), STQK(40), STQL(40), STQM(40),
40    STQN(40), STQO(40), STQP(40), STQQ(40), STQR(40), STQS(40), STQT(40), STQU(40), STQV(40), STQW(40), STQX(40),
41    STQY(40), STQZ(40), STRA(40), STRB(40), STRC(40), STRD(40), STRE(40), STRF(40), STRG(40), STRH(40), STRI(40),
42    STRJ(40), STRK(40), STRL(40), STRM(40), STRN(40), STRO(40), STRP(40), STRQ(40), STRR(40), STRS(40), STRT(40),
43    STRU(40), STRV(40), STRW(40), STRX(40), STRY(40), STRZ(40), STSA(40), STSB(40), STSC(40), STSD(40), STEB(40),
44    STSF(40), STSG(40), STSH(40), STSI(40), STSJ(40), STSK(40), STSL(40), STSM(40), STSN(40), STSO(40), STSP(40),
45    STSQ(40), STRT(40), STSS(40), STST(40), STSU(40), STSV(40), STSW(40), STSX(40), STSY(40), STSZ(40), STTA(40),
46    STTB(40), STTC(40), STTD(40), STEH(40), STTF(40), STTG(40), STTH(40), STTI(40), STTJ(40), STTK(40), STTL(40),
47    STTM(40), STTN(40), STTO(40), STTP(40), STTQ(40), STTR(40), STTS(40), STTT(40), STTU(40), STTV(40), STTW(40),
48    STTX(40), STTY(40), STTZ(40), STUA(40), STUB(40), STUC(40), STUD(40), STEH(40), STUF(40), STUG(40),
49    STUH(40), STUI(40), STUJ(40), STUK(40), STUL(40), STUM(40), STUN(40), STUO(40), STUP(40), STUQ(40), STUR(40),
50    STUS(40), STUT(40), STUU(40), STUV(40), STUW(40), STUX(40), STUY(40), STUZ(40), STVA(40), STVB(40), STVC(40),
51    STVD(40), STEB(40), STVF(40), STVG(40), STVH(40), STVI(40), STVJ(40), STVK(40), STVL(40), STVM(40), STVN(40),
52    STVO(40), STVP(40), STVQ(40), STVR(40), STVS(40), STVT(40), STVU(40), STVV(40), STVW(40), STVX(40), STVY(40),
53    STVZ(40), STWA(40), STWB(40), STWC(40), STWD(40), STEH(40), STWF(40), STWG(40), STWH(40), STWI(40), STWJ(40),
54    STWK(40), STWL(40), STWM(40), STWN(40), STWO(40), STWP(40), STWQ(40), STWR(40), STWS(40), STWT(40), STWU(40),
55    STWV(40), STWW(40), STWX(40), STWY(40), STWZ(40), STXA(40), STXB(40), STXC(40), STXD(40), STEH(40), STXF(40),
56    STXG(40), STXH(40), STXI(40), STXJ(40), STXK(40), STXL(40), STXM(40), STXN(40), STXO(40), STXP(40), STXQ(40),
57    STXR(40), STXS(40), STXT(40), STXU(40), STXV(40), STXW(40), STXX(40), STXY(40), STXZ(40), STYA(40), STYB(40),
58    STYC(40), STYD(40), STEH(40), STYF(40), STYG(40), STYH(40), STYI(40), STYJ(40), STYK(40), STYL(40), STYM(40),
59    STYN(40), STYO(40), STYP(40), STYQ(40), STYR(40), STYS(40), STYT(40), STYU(40), STYV(40), STYW(40), STYX(40),
60    STYY(40), STYZ(40), STZA(40), STZB(40), STZC(40), STZD(40), STEH(40), STZF(40), STZG(40), STZH(40), STZI(40),
61    STZJ(40), STZK(40), STZL(40), STZM(40), STZN(40), STZO(40), STZP(40), STZQ(40), STZR(40), STZS(40), STZT(40),
62    STZU(40), STZV(40), STZW(40), STZX(40), STZY(40), STZZ(40)
63     RETURN
64     END

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1 SUBROUTINE WTRMP (I)
2 C-----
3 C-----
4 C-----
5 I = I - 1
6 C-----
7 I = I + 1
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08/26/79 13.46.16

FTM 4.00030

0814C TRACE

73/74

SUBROUTINE STEEP

BT BT 102

RETURN
END

155

CARD NO - SEVERITY DETAILS DIAGNOSIS OF PROBLEM
101 I AM IS STATEMENT MAY BE MADE EFFICIENT TAB A 2 ON 2 BRANCH COMPUTED GO TO STATEMENT.

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99 2000TIME INDIENIA
100 2000TIME INDIENIA

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1106 0411=V(11)C$ALFA
1107 0412=V(11)S(17)C$D2(1)
1108 0413=SOMT(ABS(VN15)/CDS(2))
1109 0414=CG YG II(7)
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1111
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235 A(2)=ABS(1.0/11)*C(2)/D(1)
A(2)=A(2)/SQR(2)
C(2)=C(2)*SQR(2)
ATT=ATT+1
3048 SIF(1)=ABS(1.0/11)*C(1)/D(1)
D(1)=D(1)+SQR(2)
3049 SIF(1)=SIF(1)/D(1)
3050 D(1)=D(1)+SQR(2)
3051 SIF(1)=SIF(1)/D(1)
3052 SIF(1)=SIF(1)/D(1)
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3093 SIF(1)=SIF(1)/D(1)
3094 SIF(1)=SIF(1)/D(1)
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3096 SIF(1)=SIF(1)/D(1)
3097 SIF(1)=SIF(1)/D(1)
3098 SIF(1)=SIF(1)/D(1)
3099 SIF(1)=SIF(1)/D(1)
3100 SIF(1)=SIF(1)/D(1)

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1 SUBROUTINE PLOTS (MIDIM,PAR,PARISY,JPB,PARAYNES,SYMBOL  
5 C.....I,.,72.....  
10 SUBROUTINE FOR PLOTTING J CURVES OF (X(I)) AGAINST X(I).  
15 X AND Y ARE ASSUMED TO BE IN ANY RANGE EXCEPT THAT NEGATIVE VALUES  
20 ARE PLOTTED AS ZERO.  
25 P AND Y ARE SCALING FACTORS. THE RANGE J. TO L. BY DIVISION BY THE MAXIMUM,  
30 IOT IS THE VARIABLE DIMENSION FOR X.  
35 IMAK IS THE NUMBER OF Y VALUES.  
40 HARS STORES THE NAME OF THE X-AXIS.  
45 JDIM IS THE VARIABLE DIMENSION OF Y.  
50 THE ARRAY (X(I)) STORES THE X VALUES OF THE J CURVES TO BE PLOTTED.  
55 THE ARRAY (Y(I,J)) STORES THE Y VALUES OF THE J CURVES TO BE PLOTTED.  
60 .....  
65 1 DIMENSION X(IOT),Y(IJM),DIMPARS(100),DIMPAR(100),SYMBOL(I,10)  
70 2 DATA (0,1,2,3,4,5,6,7,8,9) /DIMPAR(1),DIMPAR(2),DIMPAR(3),DIMPAR(4),  
75 DIMPAR(5),DIMPAR(6),DIMPAR(7),DIMPAR(8),DIMPAR(9),DIMPAR(10) /  
80 DIMENSION X(IOT),Y(IJM),DIMPARS(100),DIMPAR(100),SYMBOL(I,10)  
85 3 IF (IOT) = 0 THEN PRINT *, 'IOT = 0' / STOP  
90 4 IF (IJM) = 0 THEN PRINT *, 'IJM = 0' / STOP  
95 5 IF (IOT) = 1 THEN PRINT *, 'IOT = 1' / STOP  
100 6 IF (IJM) = 1 THEN PRINT *, 'IJM = 1' / STOP  
105 7 IF (IOT) = 2 THEN PRINT *, 'IOT = 2' / STOP  
110 8 IF (IJM) = 2 THEN PRINT *, 'IJM = 2' / STOP  
115 9 IF (IOT) = 3 THEN PRINT *, 'IOT = 3' / STOP  
120 10 IF (IJM) = 3 THEN PRINT *, 'IJM = 3' / STOP  
125 11 IF (IOT) = 4 THEN PRINT *, 'IOT = 4' / STOP  
130 12 IF (IJM) = 4 THEN PRINT *, 'IJM = 4' / STOP  
135 13 IF (IOT) = 5 THEN PRINT *, 'IOT = 5' / STOP  
140 14 IF (IJM) = 5 THEN PRINT *, 'IJM = 5' / STOP  
145 15 IF (IOT) = 6 THEN PRINT *, 'IOT = 6' / STOP  
150 16 IF (IJM) = 6 THEN PRINT *, 'IJM = 6' / STOP  
155 17 IF (IOT) = 7 THEN PRINT *, 'IOT = 7' / STOP  
160 18 IF (IJM) = 7 THEN PRINT *, 'IJM = 7' / STOP  
165 19 IF (IOT) = 8 THEN PRINT *, 'IOT = 8' / STOP  
170 20 IF (IJM) = 8 THEN PRINT *, 'IJM = 8' / STOP  
175 21 IF (IOT) = 9 THEN PRINT *, 'IOT = 9' / STOP  
180 22 IF (IJM) = 9 THEN PRINT *, 'IJM = 9' / STOP  
185 23 IF (IOT) = 10 THEN PRINT *, 'IOT = 10' / STOP  
190 24 IF (IJM) = 10 THEN PRINT *, 'IJM = 10' / STOP  
195 25 IF (IOT) = 11 THEN PRINT *, 'IOT = 11' / STOP  
200 26 IF (IJM) = 11 THEN PRINT *, 'IJM = 11' / STOP  
205 27 IF (IOT) = 12 THEN PRINT *, 'IOT = 12' / STOP  
210 28 IF (IJM) = 12 THEN PRINT *, 'IJM = 12' / STOP  
215 29 IF (IOT) = 13 THEN PRINT *, 'IOT = 13' / STOP  
220 30 IF (IJM) = 13 THEN PRINT *, 'IJM = 13' / STOP  
225 31 IF (IOT) = 14 THEN PRINT *, 'IOT = 14' / STOP  
230 32 IF (IJM) = 14 THEN PRINT *, 'IJM = 14' / STOP  
235 33 IF (IOT) = 15 THEN PRINT *, 'IOT = 15' / STOP  
240 34 IF (IJM) = 15 THEN PRINT *, 'IJM = 15' / STOP  
245 35 IF (IOT) = 16 THEN PRINT *, 'IOT = 16' / STOP  
250 36 IF (IJM) = 16 THEN PRINT *, 'IJM = 16' / STOP  
255 37 IF (IOT) = 17 THEN PRINT *, 'IOT = 17' / STOP  
260 38 IF (IJM) = 17 THEN PRINT *, 'IJM = 17' / STOP  
265 39 IF (IOT) = 18 THEN PRINT *, 'IOT = 18' / STOP  
270 40 IF (IJM) = 18 THEN PRINT *, 'IJM = 18' / STOP  
275 41 IF (IOT) = 19 THEN PRINT *, 'IOT = 19' / STOP  
280 42 IF (IJM) = 19 THEN PRINT *, 'IJM = 19' / STOP  
285 43 IF (IOT) = 20 THEN PRINT *, 'IOT = 20' / STOP  
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295 45 IF (IOT) = 21 THEN PRINT *, 'IOT = 21' / STOP  
300 46 IF (IJM) = 21 THEN PRINT *, 'IJM = 21' / STOP  
305 47 IF (IOT) = 22 THEN PRINT *, 'IOT = 22' / STOP  
310 48 IF (IJM) = 22 THEN PRINT *, 'IJM = 22' / STOP  
315 49 IF (IOT) = 23 THEN PRINT *, 'IOT = 23' / STOP  
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325 51 IF (IOT) = 24 THEN PRINT *, 'IOT = 24' / STOP  
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345 55 IF (IOT) = 26 THEN PRINT *, 'IOT = 26' / STOP  
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355 57 IF (IOT) = 27 THEN PRINT *, 'IOT = 27' / STOP  
360 58 IF (IJM) = 27 THEN PRINT *, 'IJM = 27' / STOP  
365 59 IF (IOT) = 28 THEN PRINT *, 'IOT = 28' / STOP  
370 60 IF (IJM) = 28 THEN PRINT *, 'IJM = 28' / STOP  
375 61 IF (IOT) = 29 THEN PRINT *, 'IOT = 29' / STOP  
380 62 IF (IJM) = 29 THEN PRINT *, 'IJM = 29' / STOP  
385 63 IF (IOT) = 30 THEN PRINT *, 'IOT = 30' / STOP  
390 64 IF (IJM) = 30 THEN PRINT *, 'IJM = 30' / STOP  
395 65 IF (IOT) = 31 THEN PRINT *, 'IOT = 31' / STOP  
400 66 IF (IJM) = 31 THEN PRINT *, 'IJM = 31' / STOP  
405 67 IF (IOT) = 32 THEN PRINT *, 'IOT = 32' / STOP  
410 68 IF (IJM) = 32 THEN PRINT *, 'IJM = 32' / STOP  
415 69 IF (IOT) = 33 THEN PRINT *, 'IOT = 33' / STOP  
420 70 IF (IJM) = 33 THEN PRINT *, 'IJM = 33' / STOP  
425 71 IF (IOT) = 34 THEN PRINT *, 'IOT = 34' / STOP  
430 72 IF (IJM) = 34 THEN PRINT *, 'IJM = 34' / STOP  
435 73 IF (IOT) = 35 THEN PRINT *, 'IOT = 35' / STOP  
440 74 IF (IJM) = 35 THEN PRINT *, 'IJM = 35' / STOP  
445 75 IF (IOT) = 36 THEN PRINT *, 'IOT = 36' / STOP  
450 76 IF (IJM) = 36 THEN PRINT *, 'IJM = 36' / STOP  
455 77 IF (IOT) = 37 THEN PRINT *, 'IOT = 37' / STOP  
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465 79 IF (IOT) = 38 THEN PRINT *, 'IOT = 38' / STOP  
470 80 IF (IJM) = 38 THEN PRINT *, 'IJM = 38' / STOP  
475 81 IF (IOT) = 39 THEN PRINT *, 'IOT = 39' / STOP  
480 82 IF (IJM) = 39 THEN PRINT *, 'IJM = 39' / STOP  
485 83 IF (IOT) = 40 THEN PRINT *, 'IOT = 40' / STOP  
490 84 IF (IJM) = 40 THEN PRINT *, 'IJM = 40' / STOP  
495 85 IF (IOT) = 41 THEN PRINT *, 'IOT = 41' / STOP  
500 86 IF (IJM) = 41 THEN PRINT *, 'IJM = 41' / STOP  
505 87 IF (IOT) = 42 THEN PRINT *, 'IOT = 42' / STOP  
510 88 IF (IJM) = 42 THEN PRINT *, 'IJM = 42' / STOP  
515 89 IF (IOT) = 43 THEN PRINT *, 'IOT = 43' / STOP  
520 90 IF (IJM) = 43 THEN PRINT *, 'IJM = 43' / STOP  
525 91 IF (IOT) = 44 THEN PRINT *, 'IOT = 44' / STOP  
530 92 IF (IJM) = 44 THEN PRINT *, 'IJM = 44' / STOP  
535 93 IF (IOT) = 45 THEN PRINT *, 'IOT = 45' / STOP  
540 94 IF (IJM) = 45 THEN PRINT *, 'IJM = 45' / STOP  
545 95 IF (IOT) = 46 THEN PRINT *, 'IOT = 46' / STOP  
550 96 IF (IJM) = 46 THEN PRINT *, 'IJM = 46' / STOP  
555 97 IF (IOT) = 47 THEN PRINT *, 'IOT = 47' / STOP  
560 98 IF (IJM) = 47 THEN PRINT *, 'IJM = 47' / STOP  
565 99 IF (IOT) = 48 THEN PRINT *, 'IOT = 48' / STOP  
570 100 IF (IJM) = 48 THEN PRINT *, 'IJM = 48' / STOP  
575 101 IF (IOT) = 49 THEN PRINT *, 'IOT = 49' / STOP  
580 102 IF (IJM) = 49 THEN PRINT *, 'IJM = 49' / STOP  
585 103 IF (IOT) = 50 THEN PRINT *, 'IOT = 50' / STOP  
590 104 IF (IJM) = 50 THEN PRINT *, 'IJM = 50' / STOP  
595 105 IF (IOT) = 51 THEN PRINT *, 'IOT = 51' / STOP  
600 106 IF (IJM) = 51 THEN PRINT *, 'IJM = 51' / STOP  
605 107 IF (IOT) = 52 THEN PRINT *, 'IOT = 52' / STOP  
610 108 IF (IJM) = 52 THEN PRINT *, 'IJM = 52' / STOP  
615 109 IF (IOT) = 53 THEN PRINT *, 'IOT = 53' / STOP  
620 110 IF (IJM) = 53 THEN PRINT *, 'IJM = 53' / STOP  
625 111 IF (IOT) = 54 THEN PRINT *, 'IOT = 54' / STOP  
630 112 IF (IJM) = 54 THEN PRINT *, 'IJM = 54' / STOP  
635 113 IF (IOT) = 55 THEN PRINT *, 'IOT = 55' / STOP  
640 114 IF (IJM) = 55 THEN PRINT *, 'IJM = 55' / STOP  
645 115 IF (IOT) = 56 THEN PRINT *, 'IOT = 56' / STOP  
650 116 IF (IJM) = 56 THEN PRINT *, 'IJM = 56' / STOP  
655 117 IF (IOT) = 57 THEN PRINT *, 'IOT = 57' / STOP  
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665 119 IF (IOT) = 58 THEN PRINT *, 'IOT = 58' / STOP  
670 120 IF (IJM) = 58 THEN PRINT *, 'IJM = 58' / STOP  
675 121 IF (IOT) = 59 THEN PRINT *, 'IOT = 59' / STOP  
680 122 IF (IJM) = 59 THEN PRINT *, 'IJM = 59' / STOP  
685 123 IF (IOT) = 60 THEN PRINT *, 'IOT = 60' / STOP  
690 124 IF (IJM) = 60 THEN PRINT *, 'IJM = 60' / STOP  
695 125 IF (IOT) = 61 THEN PRINT *, 'IOT = 61' / STOP  
700 126 IF (IJM) = 61 THEN PRINT *, 'IJM = 61' / STOP  
705 127 IF (IOT) = 62 THEN PRINT *, 'IOT = 62' / STOP  
710 128 IF (IJM) = 62 THEN PRINT *, 'IJM = 62' / STOP  
715 129 IF (IOT) = 63 THEN PRINT *, 'IOT = 63' / STOP  
720 130 IF (IJM) = 63 THEN PRINT *, 'IJM = 63' / STOP  
725 131 IF (IOT) = 64 THEN PRINT *, 'IOT = 64' / STOP  
730 132 IF (IJM) = 64 THEN PRINT *, 'IJM = 64' / STOP  
735 133 IF (IOT) = 65 THEN PRINT *, 'IOT = 65' / STOP  
740 134 IF (IJM) = 65 THEN PRINT *, 'IJM = 65' / STOP  
745 135 IF (IOT) = 66 THEN PRINT *, 'IOT = 66' / STOP  
750 136 IF (IJM) = 66 THEN PRINT *, 'IJM = 66' / STOP  
755 137 IF (IOT) = 67 THEN PRINT *, 'IOT = 67' / STOP  
760 138 IF (IJM) = 67 THEN PRINT *, 'IJM = 67' / STOP  
765 139 IF (IOT) = 68 THEN PRINT *, 'IOT = 68' / STOP  
770 140 IF (IJM) = 68 THEN PRINT *, 'IJM = 68' / STOP  
775 141 IF (IOT) = 69 THEN PRINT *, 'IOT = 69' / STOP  
780 142 IF (IJM) = 69 THEN PRINT *, 'IJM = 69' / STOP  
785 143 IF (IOT) = 70 THEN PRINT *, 'IOT = 70' / STOP  
790 144 IF (IJM) = 70 THEN PRINT *, 'IJM = 70' / STOP  
795 145 IF (IOT) = 71 THEN PRINT *, 'IOT = 71' / STOP  
800 146 IF (IJM) = 71 THEN PRINT *, 'IJM = 71' / STOP  
805 147 IF (IOT) = 72 THEN PRINT *, 'IOT = 72' / STOP  
810 148 IF (IJM) = 72 THEN PRINT *, 'IJM = 72' / STOP  
815 149 IF (IOT) = 73 THEN PRINT *, 'IOT = 73' / STOP  
820 150 IF (IJM) = 73 THEN PRINT *, 'IJM = 73' / STOP  
825 151 IF (IOT) = 74 THEN PRINT *, 'IOT = 74' / STOP  
830 152 IF (IJM) = 74 THEN PRINT *, 'IJM = 74' / STOP  
835 153 IF (IOT) = 75 THEN PRINT *, 'IOT = 75' / STOP  
840 154 IF (IJM) = 75 THEN PRINT *, 'IJM = 75' / STOP  
845 155 IF (IOT) = 76 THEN PRINT *, 'IOT = 76' / STOP  
850 156 IF (IJM) = 76 THEN PRINT *, 'IJM = 76' / STOP  
855 157 IF (IOT) = 77 THEN PRINT *, 'IOT = 77' / STOP  
860 158 IF (IJM) = 77 THEN PRINT *, 'IJM = 77' / STOP  
865 159 IF (IOT) = 78 THEN PRINT *, 'IOT = 78' / STOP  
870 160 IF (IJM) = 78 THEN PRINT *, 'IJM = 78' / STOP  
875 161 IF (IOT) = 79 THEN PRINT *, 'IOT = 79' / STOP  
880 162 IF (IJM) = 79 THEN PRINT *, 'IJM = 79' / STOP  
885 163 IF (IOT) = 80 THEN PRINT *, 'IOT = 80' / STOP  
890 164 IF (IJM) = 80 THEN PRINT *, 'IJM = 80' / STOP  
895 165 IF (IOT) = 81 THEN PRINT *, 'IOT = 81' / STOP  
900 166 IF (IJM) = 81 THEN PRINT *, 'IJM = 81' / STOP  
905 167 IF (IOT) = 82 THEN PRINT *, 'IOT = 82' / STOP  
910 168 IF (IJM) = 82 THEN PRINT *, 'IJM = 82' / STOP  
915 169 IF (IOT) = 83 THEN PRINT *, 'IOT = 83' / STOP  
920 170 IF (IJM) = 83 THEN PRINT *, 'IJM = 83' / STOP  
925 171 IF (IOT) = 84 THEN PRINT *, 'IOT = 84' / STOP  
930 172 IF (IJM) = 84 THEN PRINT *, 'IJM = 84' / STOP  
935 173 IF (IOT) = 85 THEN PRINT *, 'IOT = 85' / STOP  
940 174 IF (IJM) = 85 THEN PRINT *, 'IJM = 85' / STOP  
945 175 IF (IOT) = 86 THEN PRINT *, 'IOT = 86' / STOP  
950 176 IF (IJM) = 86 THEN PRINT *, 'IJM = 86' / STOP  
955 177 IF (IOT) = 87 THEN PRINT *, 'IOT = 87' / STOP  
960 178 IF (IJM) = 87 THEN PRINT *, 'IJM = 87' / STOP  
965 179 IF (IOT) = 88 THEN PRINT *, 'IOT = 88' / STOP  
970 180 IF (IJM) = 88 THEN PRINT *, 'IJM = 88' / STOP  
975 181 IF (IOT) = 89 THEN PRINT *, 'IOT = 89' / STOP  
980 182 IF (IJM) = 89 THEN PRINT *, 'IJM = 89' / STOP  
985 183 IF (IOT) = 90 THEN PRINT *, 'IOT = 90' / STOP  
990 184 IF (IJM) = 90 THEN PRINT *, 'IJM = 90' / STOP  
995 185 IF (IOT) = 91 THEN PRINT *, 'IOT = 91' / STOP  
1000 186 IF (IJM) = 91 THEN PRINT *, 'IJM = 91' / STOP
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APPENDIX E

LISTING OF TRANSITION MIXING MODEL

PROGRAM MAIN 73/74 OPT=0 TRACE FTM 4-84439 08/24/79 10.30.04 PAGE 1

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1 PROGRAM MAIN 1 INPUT, OUTPUT, TAPE=INPUT, TAPE=OUTPUT
5 TRANSITION MIXING MODEL.
10 COMMON/CDNA/AY(3), AE(7), AJ(7), B(43), CL(3), CS(7), D(43), DPMX(43),
11 VTE(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
15 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
20 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
25 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
30 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
35 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
40 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
45 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
50 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
55 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
60 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
65 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
70 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),
75 S(7), T(7), U(7), V(7), W(7), X(7), Y(7), Z(7), M(7), N(7), O(7), P(7), Q(7), R(7),

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00 C 14 FORMAT (6E12.6) Z-VALUES
01 C 15 CALL MATR(Z)
02 C 16 DO 17 I=1,N
03 C 17 DO 18 J=1,N
04 C 18 DO 19 K=1,N
05 C 19 DO 20 L=1,N
06 C 20 DO 21 M=1,N
07 C 21 DO 22 P=1,N
08 C 22 DO 23 Q=1,N
09 C 23 DO 24 R=1,N
10 C 24 DO 25 S=1,N
11 C 25 DO 26 T=1,N
12 C 26 DO 27 U=1,N
13 C 27 DO 28 V=1,N
14 C 28 DO 29 W=1,N
15 C 29 DO 30 X=1,N
16 C 30 DO 31 Y=1,N
17 C 31 DO 32 Z=1,N
18 C 32 DO 33 AA=1,N
19 C 33 DO 34 AB=1,N
20 C 34 DO 35 AC=1,N
21 C 35 DO 36 AD=1,N
22 C 36 DO 37 AE=1,N
23 C 37 DO 38 AF=1,N
24 C 38 DO 39 AG=1,N
25 C 39 DO 40 AH=1,N
26 C 40 DO 41 AI=1,N
27 C 41 DO 42 AJ=1,N
28 C 42 DO 43 AK=1,N
29 C 43 DO 44 AL=1,N
30 C 44 DO 45 AM=1,N
31 C 45 DO 46 AN=1,N
32 C 46 DO 47 AO=1,N
33 C 47 DO 48 AP=1,N
34 C 48 DO 49 AQ=1,N
35 C 49 DO 50 AR=1,N
36 C 50 DO 51 AS=1,N
37 C 51 DO 52 AT=1,N
38 C 52 DO 53 AU=1,N
39 C 53 DO 54 AV=1,N
40 C 54 DO 55 AW=1,N
41 C 55 DO 56 AX=1,N
42 C 56 DO 57 AY=1,N
43 C 57 DO 58 AZ=1,N
44 C 58 DO 59 BA=1,N
45 C 59 DO 60 BB=1,N
46 C 60 DO 61 BC=1,N
47 C 61 DO 62 BD=1,N
48 C 62 DO 63 BE=1,N
49 C 63 DO 64 BF=1,N
50 C 64 DO 65 BG=1,N
51 C 65 DO 66 BH=1,N
52 C 66 DO 67 BI=1,N
53 C 67 DO 68 BJ=1,N
54 C 68 DO 69 BK=1,N
55 C 69 DO 70 BL=1,N
56 C 70 DO 71 BM=1,N
57 C 71 DO 72 BN=1,N
58 C 72 DO 73 BO=1,N
59 C 73 DO 74 BP=1,N
60 C 74 DO 75 BQ=1,N
61 C 75 DO 76 BR=1,N
62 C 76 DO 77 BS=1,N
63 C 77 DO 78 BT=1,N
64 C 78 DO 79 BU=1,N
65 C 79 DO 80 BV=1,N
66 C 80 DO 81 BW=1,N
67 C 81 DO 82 BX=1,N
68 C 82 DO 83 BY=1,N
69 C 83 DO 84 BZ=1,N
70 C 84 DO 85 CA=1,N
71 C 85 DO 86 CB=1,N
72 C 86 DO 87 CC=1,N
73 C 87 DO 88 CD=1,N
74 C 88 DO 89 CE=1,N
75 C 89 DO 90 CF=1,N
76 C 90 DO 91 CG=1,N
77 C 91 DO 92 CH=1,N
78 C 92 DO 93 CI=1,N
79 C 93 DO 94 CJ=1,N
80 C 94 DO 95 CK=1,N
81 C 95 DO 96 CL=1,N
82 C 96 DO 97 CM=1,N
83 C 97 DO 98 CN=1,N
84 C 98 DO 99 CO=1,N
85 C 99 DO 100 CP=1,N
86 C 100 DO 101 CQ=1,N
87 C 101 DO 102 CR=1,N
88 C 102 DO 103 CS=1,N
89 C 103 DO 104 CT=1,N
90 C 104 DO 105 CU=1,N
91 C 105 DO 106 CV=1,N
92 C 106 DO 107 CW=1,N
93 C 107 DO 108 CX=1,N
94 C 108 DO 109 CY=1,N
95 C 109 DO 110 CZ=1,N
96 C 110 DO 111 DA=1,N
97 C 111 DO 112 DB=1,N
98 C 112 DO 113 DC=1,N
99 C 113 DO 114 DD=1,N
100 C 114 DO 115 DE=1,N
101 C 115 DO 116 DF=1,N
102 C 116 DO 117 DG=1,N
103 C 117 DO 118 DH=1,N
104 C 118 DO 119 DI=1,N
105 C 119 DO 120 DJ=1,N
106 C 120 DO 121 DK=1,N
107 C 121 DO 122 DL=1,N
108 C 122 DO 123 DM=1,N
109 C 123 DO 124 DN=1,N
110 C 124 DO 125 DO=1,N
111 C 125 DO 126 DP=1,N
112 C 126 DO 127 DQ=1,N
113 C 127 DO 128 DR=1,N
114 C 128 DO 129 DS=1,N
115 C 129 DO 130 DT=1,N
116 C 130 DO 131 DU=1,N
117 C 131 DO 132 DV=1,N
118 C 132 DO 133 DW=1,N
119 C 133 DO 134 DX=1,N
120 C 134 DO 135 DY=1,N
121 C 135 DO 136 DZ=1,N
122 C 136 DO 137 EA=1,N
123 C 137 DO 138 EB=1,N
124 C 138 DO 139 EC=1,N
125 C 139 DO 140 ED=1,N
126 C 140 DO 141 EE=1,N
127 C 141 DO 142 EF=1,N
128 C 142 DO 143 EG=1,N
129 C 143 DO 144 EH=1,N
130 C 144 DO 145 EI=1,N
131 C 145 DO 146 EJ=1,N
132 C 146 DO 147 EK=1,N
133 C 147 DO 148 EL=1,N
134 C 148 DO 149 EM=1,N
135 C 149 DO 150 EN=1,N
136 C 150 DO 151 EO=1,N
137 C 151 DO 152 EP=1,N
138 C 152 DO 153 EQ=1,N
139 C 153 DO 154 ER=1,N
140 C 154 DO 155 ES=1,N
141 C 155 DO 156 ET=1,N
142 C 156 DO 157 EU=1,N
143 C 157 DO 158 EV=1,N
144 C 158 DO 159 EW=1,N
145 C 159 DO 160 EX=1,N
146 C 160 DO 161 EY=1,N
147 C 161 DO 162 EZ=1,N
148 C 162 DO 163 FA=1,N
149 C 163 DO 164 FB=1,N
150 C 164 DO 165 FC=1,N

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310 64 DO 10 I=1,MP3
311 65 1000,600, I=INT
312 66 1000,600, I=INT
313 67 1000,600, I=INT
314 68 1000,600, I=INT
315 69 1000,600, I=INT
316 70 1000,600, I=INT
317 71 1000,600, I=INT
318 72 1000,600, I=INT
319 73 1000,600, I=INT
320 74 1000,600, I=INT
321 75 1000,600, I=INT
322 76 1000,600, I=INT
323 77 1000,600, I=INT
324 78 1000,600, I=INT
325 79 1000,600, I=INT
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327 81 1000,600, I=INT
328 82 1000,600, I=INT
329 83 1000,600, I=INT
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334 88 1000,600, I=INT
335 89 1000,600, I=INT
336 90 1000,600, I=INT
337 91 1000,600, I=INT
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339 93 1000,600, I=INT
340 94 1000,600, I=INT
341 95 1000,600, I=INT
342 96 1000,600, I=INT
343 97 1000,600, I=INT
344 98 1000,600, I=INT
345 99 1000,600, I=INT
346 100 1000,600, I=INT
347 101 1000,600, I=INT
348 102 1000,600, I=INT
349 103 1000,600, I=INT
350 104 1000,600, I=INT
351 105 1000,600, I=INT
352 106 1000,600, I=INT
353 107 1000,600, I=INT
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357 111 1000,600, I=INT
358 112 1000,600, I=INT
359 113 1000,600, I=INT
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389 143 1000,600, I=INT
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392 146 1000,600, I=INT
393 147 1000,600, I=INT
394 148 1000,600, I=INT
395 149 1000,600, I=INT
396 150 1000,600, I=INT
397 151 1000,600, I=INT
398 152 1000,600, I=INT
399 153 1000,600, I=INT
400 154 1000,600, I=INT

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465      BR=STEP-EG-0) TAVE=0.
470      82  A2E(1)=0
475      75  I=I+1
480      75  I=I+1
485      75  I=I+1
490      75  I=I+1
495      75  I=I+1
500      75  I=I+1
505      75  I=I+1
510      75  I=I+1
515      75  I=I+1
520      75  I=I+1
525      75  I=I+1
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535      75  I=I+1
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595      75  I=I+1
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605      75  I=I+1
610      75  I=I+1
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680      75  I=I+1
685      75  I=I+1
690      75  I=I+1
695      75  I=I+1
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765      75  I=I+1
770      75  I=I+1
775      75  I=I+1
780      75  I=I+1
785      75  I=I+1
790      75  I=I+1
795      75  I=I+1
800      75  I=I+1
805      75  I=I+1
810      75  I=I+1
815      75  I=I+1
820      75  I=I+1
825      75  I=I+1
830      75  I=I+1
835      75  I=I+1
840      75  I=I+1
845      75  I=I+1
850      75  I=I+1
855      75  I=I+1
860      75  I=I+1
865      75  I=I+1
870      75  I=I+1
875      75  I=I+1
880      75  I=I+1
885      75  I=I+1
890      75  I=I+1
895      75  I=I+1
900      75  I=I+1
905      75  I=I+1
910      75  I=I+1
915      75  I=I+1
920      75  I=I+1
925      75  I=I+1
930      75  I=I+1
935      75  I=I+1
940      75  I=I+1
945      75  I=I+1
950      75  I=I+1
955      75  I=I+1
960      75  I=I+1
965      75  I=I+1
970      75  I=I+1
975      75  I=I+1
980      75  I=I+1
985      75  I=I+1
990      75  I=I+1
995      75  I=I+1

```


PROGRAM MAIN 73/74 OPT=0 TRACE
DIAGNOSIS OF PROBLER

CARD NR. SEVERITY DETAILS
I I
I I
I I
I I

310 AN IF STATEMENT MAY BE MORE EFFICIENT THAN 4 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
320 AN IF STATEMENT MAY BE MORE EFFICIENT THAN 4 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
330 AN IF STATEMENT MAY BE MORE EFFICIENT THAN 4 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
340 AN IF STATEMENT MAY BE MORE EFFICIENT THAN 4 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
350 AN IF STATEMENT MAY BE MORE EFFICIENT THAN 4 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.

FUNCTION RCUW 7374 OPT=0 TRACE

```

1 FUNCTION RCUW (Z1, Z2, Y, Y1)
COMMON/CUR/MCUE, Z1, Z2, Y1, Y, ZR1(40), ZR2(40), RRI(40),
10 MCV1(30), RCV1(30)
11 RCUW=Z1*(RRI(40)+Y*(ZR1(40)+Z2*(ZR2(40)-ZR1(40))))/
5 C ..... INTERNAL BOUNDARY RADIUS OF CURVATURE .....
MCV1+Y*(RRI(40)+Z2*(RRI(40)+Z1*(RRI(40)-RRI(40))))/
GO TO 15
10 C ..... EXTERNAL BOUNDARY RADIUS OF CURVATURE .....
15 RCUW=MCV1*(RRI(40)+Y*(ZR1(40)+Z2*(ZR2(40)-ZR1(40))))/
GO TO 20
15 C ..... ACTUAL RCU VALUE IS LINEAR INTERPOLATION .....
20 RCUW=RRI*(MCV1+Y*(Z1*(RRI(40)-RRI(40))))
END

```

RC
CURC
CURC
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```

1  FUNCTION TAB (I,XI,YY,M*)
5  DIMENSION TAB(10,20,YY,M*)
   TAB=0
   RETURN
10  IF (I) GO TO 15
   IF (I) GO TO 15
   IF (I) GO TO 15
15  IF (I) GO TO 15
   IF (I) GO TO 15
   IF (I) GO TO 15
20  IF (I) GO TO 15
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30  IF (I) GO TO 15
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   IF (I) GO TO 15
40  IF (I) GO TO 15
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   IF (I) GO TO 15
50  IF (I) GO TO 15
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   IF (I) GO TO 15
60  IF (I) GO TO 15
   IF (I) GO TO 15
   IF (I) GO TO 15
END
    
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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM
 7 I XI ARRAY REFERENCE OUTSIDE DIMENSION COUNTS.


```
235 C QV10Y-(V13-VT2)/R2Z  
R2Z=(R23-RK2)/R2Z  
CEM(I2)-OUBV(I2)+2*(R10VTY-VT2)/(.75+R(I2)+.25+R(I3))00Z  
R00 J00 I=3.MP1  
R22-V(I)+11-V(I)  
IF (.75+R(I2)) R2Z=V(MP1)-V(MP1)  
R10Y=V(I)+OZZ  
R11=V(I)-OZZ  
R12=V(I)  
V13=JVT, I=11/M(I)  
IF (.75+R(I2)) V13=JVT+MP2/(.75+R(MP2)+.25+R(MP1))  
R10Y=V(I)+V(I)-V(I)+V(I)+OZZ  
R12=V(I)  
R13=V(I)  
R14=V(I)  
R15=V(I)  
R16=V(I)  
R17=V(I)  
R18=V(I)  
R19=V(I)  
R20=V(I)  
R21=V(I)  
R22=V(I)  
R23=V(I)  
R24=V(I)  
R25=V(I)  
R26=V(I)  
R27=V(I)  
R28=V(I)  
R29=V(I)  
R30=V(I)  
R31=V(I)  
R32=V(I)  
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R40=V(I)  
R41=V(I)  
R42=V(I)  
R43=V(I)  
R44=V(I)  
R45=V(I)  
R46=V(I)  
R47=V(I)  
R48=V(I)  
R49=V(I)  
R50=V(I)
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```

1 SUBROUTINE STRIDE(ISH)
2 COMMON/COMA/ALF(3),AD(2),AJ(2),B(3),CSALFA(6),C(4),D(3),E(3),F(3),G(3),H(3),I(3),
3 J(3),K(3),L(3),M(3),N(3),O(3),P(3),Q(3),R(3),S(3),T(3),U(3),V(3),W(3),X(3),Y(3),Z(3),
4 AA(3),AB(3),AC(3),AD(3),AE(3),AF(3),AG(3),AH(3),AI(3),AJ(3),AK(3),AL(3),AM(3),AN(3),
5 AO(3),AP(3),AQ(3),AR(3),AS(3),AT(3),AU(3),AV(3),AW(3),AX(3),AY(3),AZ(3),BA(3),
6 BB(3),BC(3),BD(3),BE(3),BF(3),BG(3),BH(3),BI(3),BJ(3),BK(3),BL(3),BM(3),BN(3),
7 BO(3),BP(3),BQ(3),BR(3),BS(3),BT(3),BU(3),BV(3),BW(3),BX(3),BY(3),BZ(3),CA(3),
8 CB(3),CC(3),CD(3),CE(3),CF(3),CG(3),CH(3),CI(3),CJ(3),CK(3),CL(3),CM(3),CN(3),
9 CO(3),CP(3),CQ(3),CR(3),CS(3),CT(3),CU(3),CV(3),CW(3),CX(3),CY(3),CZ(3),DA(3),
10 DB(3),DC(3),DD(3),DE(3),DF(3),DG(3),DH(3),DI(3),DJ(3),DK(3),DL(3),DM(3),DN(3),
11 DO(3),DP(3),DQ(3),DR(3),DS(3),DT(3),DU(3),DV(3),DW(3),DX(3),DY(3),DZ(3),EA(3),
12 EB(3),EC(3),ED(3),EE(3),EF(3),EG(3),EH(3),EI(3),EJ(3),EK(3),EL(3),EM(3),EN(3),
13 EO(3),EP(3),EQ(3),ER(3),ES(3),ET(3),EU(3),EV(3),EW(3),EX(3),EY(3),EZ(3),FA(3),
14 FB(3),FC(3),FD(3),FE(3),FF(3),FG(3),FH(3),FI(3),FJ(3),FK(3),FL(3),FM(3),FN(3),
15 FO(3),FP(3),FQ(3),FR(3),FS(3),FT(3),FU(3),FV(3),FW(3),FX(3),FY(3),FZ(3),GA(3),
16 GB(3),GC(3),GD(3),GE(3),GF(3),GG(3),GH(3),GI(3),GJ(3),GK(3),GL(3),GM(3),GN(3),
17 GO(3),GP(3),GQ(3),GR(3),GS(3),GT(3),GU(3),GV(3),GW(3),GX(3),GY(3),GZ(3),HA(3),
18 HB(3),HC(3),HD(3),HE(3),HF(3),HG(3),HH(3),HI(3),HJ(3),HK(3),HL(3),HM(3),HN(3),
19 HO(3),HP(3),HQ(3),HR(3),HS(3),HT(3),HU(3),HV(3),HW(3),HX(3),HY(3),HZ(3),IA(3),
20 IB(3),IC(3),ID(3),IE(3),IF(3),IG(3),IH(3),II(3),IJ(3),IK(3),IL(3),IM(3),IN(3),
21 IO(3),IP(3),IQ(3),IR(3),IS(3),IT(3),IU(3),IV(3),IW(3),IX(3),IY(3),IZ(3),JA(3),
22 JB(3),JC(3),JD(3),JE(3),JF(3),JG(3),JH(3),JI(3),JJ(3),JK(3),JL(3),JM(3),JN(3),
23 JO(3),JP(3),JQ(3),JR(3),JS(3),JT(3),JU(3),JV(3),JW(3),JX(3),JY(3),JZ(3),KA(3),
24 KB(3),KC(3),KD(3),KE(3),KF(3),KG(3),KH(3),KI(3),KJ(3),KK(3),KL(3),KM(3),KN(3),
25 KO(3),KP(3),KQ(3),KR(3),KS(3),KT(3),KU(3),KV(3),KW(3),KX(3),KY(3),KZ(3),LA(3),
26 LB(3),LC(3),LD(3),LE(3),LF(3),LG(3),LH(3),LI(3),LJ(3),LK(3),LL(3),LM(3),LN(3),
27 LO(3),LP(3),LQ(3),LR(3),LS(3),LT(3),LU(3),LV(3),LW(3),LX(3),LY(3),LZ(3),MA(3),
28 MB(3),MC(3),MD(3),ME(3),MF(3),MG(3),MH(3),MI(3),MJ(3),MK(3),ML(3),MM(3),MN(3),
29 MO(3),MP(3),MQ(3),MR(3),MS(3),MT(3),MU(3),MV(3),MW(3),MX(3),MY(3),MZ(3),NA(3),
30 NB(3),NC(3),ND(3),NE(3),NF(3),NG(3),NH(3),NI(3),NJ(3),NK(3),NL(3),NM(3),NO(3),
31 NP(3),NQ(3),NR(3),NS(3),NT(3),NU(3),NV(3),NW(3),NX(3),NY(3),NZ(3),OA(3),
32 OB(3),OC(3),OD(3),OE(3),OF(3),OG(3),OH(3),OI(3),OJ(3),OK(3),OL(3),OM(3),ON(3),
33 OO(3),OP(3),OQ(3),OR(3),OS(3),OT(3),OU(3),OV(3),OW(3),OX(3),OY(3),OZ(3),PA(3),
34 PB(3),PC(3),PD(3),PE(3),PF(3),PG(3),PH(3),PI(3),PJ(3),PK(3),PL(3),PM(3),PN(3),
35 PO(3),PP(3),PQ(3),PR(3),PS(3),PT(3),PU(3),PV(3),PW(3),PX(3),PY(3),PZ(3),QA(3),
36 QB(3),QC(3),QD(3),QE(3),QF(3),QG(3),QH(3),QI(3),QJ(3),QK(3),QL(3),QM(3),QN(3),
37 QO(3),QP(3),QQ(3),QR(3),QS(3),QT(3),QU(3),QV(3),QW(3),QX(3),QY(3),QZ(3),RA(3),
38 RB(3),RC(3),RD(3),RE(3),RF(3),RG(3),RH(3),RI(3),RJ(3),RK(3),RL(3),RM(3),RN(3),
39 RO(3),RP(3),RQ(3),RR(3),RS(3),RT(3),RU(3),RV(3),RW(3),RX(3),RY(3),RZ(3),SA(3),
40 SB(3),SC(3),SD(3),SE(3),SF(3),SG(3),SH(3),SI(3),SJ(3),SK(3),SL(3),SM(3),SN(3),
41 SO(3),SP(3),SQ(3),SR(3),SS(3),ST(3),SU(3),SV(3),SW(3),SX(3),SY(3),SZ(3),TA(3),
42 TB(3),TC(3),TD(3),TE(3),TF(3),TG(3),TH(3),TI(3),TJ(3),TK(3),TL(3),TM(3),TN(3),
43 TO(3),TP(3),TQ(3),TR(3),TS(3),TT(3),TU(3),TV(3),TW(3),TX(3),TY(3),TZ(3),UA(3),
44 UB(3),UC(3),UD(3),UE(3),UF(3),UG(3),UH(3),UI(3),UJ(3),UK(3),UL(3),UM(3),UN(3),
45 UO(3),UP(3),UQ(3),UR(3),US(3),UT(3),UU(3),UV(3),UW(3),UX(3),UY(3),UZ(3),VA(3),
46 VB(3),VC(3),VD(3),VE(3),VF(3),VG(3),VH(3),VI(3),VJ(3),VK(3),VL(3),VM(3),VN(3),
47 VO(3),VP(3),VQ(3),VR(3),VS(3),VT(3),VU(3),VV(3),VW(3),VX(3),VY(3),VZ(3),WA(3),
48 WB(3),WC(3),WD(3),WE(3),WF(3),WG(3),WH(3),WI(3),WJ(3),WK(3),WL(3),WM(3),WN(3),
49 WO(3),WP(3),WQ(3),WR(3),WS(3),WT(3),WU(3),WV(3),WW(3),WX(3),WY(3),WZ(3),XA(3),
50 XB(3),XC(3),XD(3),XE(3),XF(3),XG(3),XH(3),XI(3),XJ(3),XK(3),XL(3),XM(3),XN(3),
51 XO(3),XP(3),XQ(3),XR(3),XS(3),XT(3),XU(3),XV(3),XW(3),XX(3),XY(3),XZ(3),YA(3),
52 YB(3),YC(3),YD(3),YE(3),YF(3),YG(3),YH(3),YI(3),YJ(3),YK(3),YL(3),YM(3),YN(3),
53 YO(3),YP(3),YQ(3),YR(3),YS(3),YT(3),YU(3),YV(3),YW(3),YX(3),YY(3),YZ(3),ZA(3),
54 ZB(3),ZC(3),ZD(3),ZE(3),ZF(3),ZG(3),ZH(3),ZI(3),ZJ(3),ZK(3),ZL(3),ZM(3),ZN(3),
55 ZO(3),ZP(3),ZQ(3),ZR(3),ZS(3),ZT(3),ZU(3),ZV(3),ZW(3),ZX(3),ZY(3),ZZ(3)
56 DATA IZAT/1.1,1.1,0.9/
57
58 C GO TO (1000,2000,3000,4000), ISH
59
60 C-----
61 C-----
62 C-----
63 C-----
64 C-----
65 C-----
66 C-----
67 C-----
68 C-----
69 C-----
70 C-----
71 C-----
72 C-----
73 C-----

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235 V1=1, N=20
 C 3050 CONST=0
 IF (M=0) GOTO 3011
 IF (M=1) GOTO 3012
 IF (M=2) GOTO 3013
 IF (M=3) GOTO 3014
 IF (M=4) GOTO 3015
 IF (M=5) GOTO 3016
 IF (M=6) GOTO 3017
 IF (M=7) GOTO 3018
 IF (M=8) GOTO 3019
 IF (M=9) GOTO 3020
 IF (M=10) GOTO 3021
 IF (M=11) GOTO 3022
 IF (M=12) GOTO 3023
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 IF (M=38) GOTO 3049
 IF (M=39) GOTO 3050
 IF (M=40) GOTO 3051
 IF (M=41) GOTO 3052
 IF (M=42) GOTO 3053
 IF (M=43) GOTO 3054
 IF (M=44) GOTO 3055
 IF (M=45) GOTO 3056
 IF (M=46) GOTO 3057
 IF (M=47) GOTO 3058
 IF (M=48) GOTO 3059
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 IF (M=82) GOTO 3093
 IF (M=83) GOTO 3094
 IF (M=84) GOTO 3095
 IF (M=85) GOTO 3096
 IF (M=86) GOTO 3097
 IF (M=87) GOTO 3098
 IF (M=88) GOTO 3099
 IF (M=89) GOTO 3100
 IF (M=90) GOTO 3101
 IF (M=91) GOTO 3102

SUBROUTINE STRIDE 73/74 OPT=0 TRACE

PAGE 6

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MPZ=99
MPZ=100

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CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM

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390 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
391 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
392 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
393 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
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399 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
400 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
401 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
402 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
403 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
404 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.
405 I AN IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.

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1  SUBROUTINE PLOTS (L,IOIR,IMAX,IAIS,IOI,IMAXI,IAISI,SYMBOL)
C.....
2  SUBROUTINE FOR PLOTTING J CURVES OF Y(I,J) AGAINST X(I).
3  X AND Y ARE ASSUMED TO BE IN ANY RANGE EXCEPT THAT NEGATIVE VALUES
4  MUST BE SCALED TO THE RANGE 0. TO 1. BY DIVISION BY THE MAXIMUM
5  WHICH ARE ALSO PRINTED.
6  IOIR IS THE VARIABLE DIMENSION FOR X.
7  IOI IS THE NUMBER OF X VALUES.
8  IMAXI IS THE NUMBER OF Y VALUES.
9  IAISI IS THE NUMBER OF CURVES TO BE PLOTTED. UP TO 101.
10  THE ARRAY YAES(I,J) STORES THE VALUES OF THE CURVES.
11  THE ARRAY SYM(I,J) STORES THE SINGLE CHARACTERS USED FOR PLOTTING.
C.....
12  DIMENSION IOI(1),YDIM(1:IMAXI),YRES(1:IMAXI),SYM(1:IOI,1)
13  DATA /BLANK/ (0),BLANK/IMAXI+1M /
14  C..... SEALING X ARRAY TO THE RANGE 0 TO 50
15  XN=1.0E-30
16  DO 1 I=1,IOI
17  X(I)=X(I)/XN
18  1 CONTINUE
19  C.....
20  DO 2 I=1,IMAXI
21  Y(I)=Y(I)/YRES(I)
22  2 CONTINUE
23  C..... SCALING Y ARRAY TO THE RANGE 0 TO 100
24  DN=1.0E-10
25  DO 3 J=1,IMAXI
26  Y(J)=Y(J)/DN
27  3 CONTINUE
28  C.....
29  DO 4 I=1,IOI
30  YMAX(I)=Y(I,IMAXI)
31  4 CONTINUE
32  DO 5 I=1,IOI
33  YMIN(I)=Y(I,1)
34  5 CONTINUE
35  DO 6 I=1,IOI
36  YRANGE(I)=(YMAX(I)-YMIN(I))/YMAX(I)
37  6 CONTINUE
38  DO 7 I=1,IOI
39  YRANGE(I)=MAX(YRANGE(I),.1)
40  7 CONTINUE
41  DO 8 I=1,IOI
42  YRANGE(I)=MIN(YRANGE(I),1.0)
43  8 CONTINUE
44  DO 9 I=1,IOI
45  YRANGE(I)=1./YRANGE(I)
46  9 CONTINUE
47  DO 10 I=1,IOI
48  YRANGE(I)=MIN(YRANGE(I),1.0)
49  10 CONTINUE
50  ALWAYE = 'R' + AS MARKER ON THE Y-AXIS
51  DO 11 I=1,IOI
52  ALWAYE(I)=ALWAYE
53  11 CONTINUE
54  ALWAYE = 'P' + MARK ON THE X-AXIS, ALSO THE APPROPRIATE F VALUE
55  DO 12 I=1,IOI
56  ALWAYE(I)=ALWAYE
57  12 CONTINUE
58  DO 13 I=1,IOI
59  ALWAYE(I)=ALWAYE
60  13 CONTINUE
61  DO 14 I=1,IOI
62  ALWAYE(I)=ALWAYE
63  14 CONTINUE
64  DO 15 I=1,IOI
65  ALWAYE(I)=ALWAYE
66  15 CONTINUE
67  DO 16 I=1,IOI
68  ALWAYE(I)=ALWAYE
69  16 CONTINUE
70  DO 17 I=1,IOI
71  ALWAYE(I)=ALWAYE
72  17 CONTINUE
73  DO 18 I=1,IOI
74  ALWAYE(I)=ALWAYE
75  18 CONTINUE
76  DO 19 I=1,IOI
77  ALWAYE(I)=ALWAYE
78  19 CONTINUE

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49 ALG1-BLANK
48 CONTINUE
50 50 1.2111
51 1.2111 1.2111
52 1.2111 1.2111 (ALG1,1-1.211)
53 1.2111 1.2111
100 FORMAT(1X, V=ARE, ABEPS,10(1X,AB,0.44))
101 FORMAT(100, 2X,10,0.11)
102 FORMAT(1X, A=10M, 1X, VALS,2,PIE(1,2))
103 FORMAT(1X, A=10M, 1X, AB, 1M, MAXIMUM VALUE =,1PE10.3)
104 FORMAT(2X, 1.211, 2X, 1.101A,2A,2)
105 FORMAT(7X, SYMOL,11,10(1X,010))
106 END
  
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155 IPRMTC(1)=IMD(1)+TEMP
156 IPRMTC(2)=IMD(2)+TEMP
157 IPRMTC(3)=IMD(3)+TEMP
158 IPRMTC(4)=IMD(4)+TEMP
159 IPRMTC(5)=IMD(5)+TEMP
160 IPRMTC(6)=IMD(6)+TEMP
161 IPRMTC(7)=IMD(7)+TEMP
162 IPRMTC(8)=IMD(8)+TEMP
163 IPRMTC(9)=IMD(9)+TEMP
164 IPRMTC(10)=IMD(10)+TEMP
165 IPRMTC(11)=IMD(11)+TEMP
166 IPRMTC(12)=IMD(12)+TEMP
167 IPRMTC(13)=IMD(13)+TEMP
168 IPRMTC(14)=IMD(14)+TEMP
169 IPRMTC(15)=IMD(15)+TEMP
170 IPRMTC(16)=IMD(16)+TEMP
171 IPRMTC(17)=IMD(17)+TEMP
172 IPRMTC(18)=IMD(18)+TEMP
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174 IPRMTC(20)=IMD(20)+TEMP
175 IPRMTC(21)=IMD(21)+TEMP
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178 IPRMTC(24)=IMD(24)+TEMP
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181 IPRMTC(27)=IMD(27)+TEMP
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183 IPRMTC(29)=IMD(29)+TEMP
184 IPRMTC(30)=IMD(30)+TEMP
185 IPRMTC(31)=IMD(31)+TEMP
186 IPRMTC(32)=IMD(32)+TEMP
187 IPRMTC(33)=IMD(33)+TEMP
188 IPRMTC(34)=IMD(34)+TEMP
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191 IPRMTC(37)=IMD(37)+TEMP
192 IPRMTC(38)=IMD(38)+TEMP
193 IPRMTC(39)=IMD(39)+TEMP
194 IPRMTC(40)=IMD(40)+TEMP
195 IPRMTC(41)=IMD(41)+TEMP
196 IPRMTC(42)=IMD(42)+TEMP
197 IPRMTC(43)=IMD(43)+TEMP
198 IPRMTC(44)=IMD(44)+TEMP
199 IPRMTC(45)=IMD(45)+TEMP
200 IPRMTC(46)=IMD(46)+TEMP
201 IPRMTC(47)=IMD(47)+TEMP
202 IPRMTC(48)=IMD(48)+TEMP
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223 IPRMTC(69)=IMD(69)+TEMP
224 IPRMTC(70)=IMD(70)+TEMP
225 IPRMTC(71)=IMD(71)+TEMP
226 IPRMTC(72)=IMD(72)+TEMP
227 IPRMTC(73)=IMD(73)+TEMP
228 IPRMTC(74)=IMD(74)+TEMP
229 IPRMTC(75)=IMD(75)+TEMP
230 IPRMTC(76)=IMD(76)+TEMP
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232 IPRMTC(78)=IMD(78)+TEMP
233 IPRMTC(79)=IMD(79)+TEMP
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235 IPRMTC(81)=IMD(81)+TEMP
236 IPRMTC(82)=IMD(82)+TEMP
237 IPRMTC(83)=IMD(83)+TEMP
238 IPRMTC(84)=IMD(84)+TEMP
239 IPRMTC(85)=IMD(85)+TEMP
240 IPRMTC(86)=IMD(86)+TEMP
241 IPRMTC(87)=IMD(87)+TEMP
242 IPRMTC(88)=IMD(88)+TEMP
243 IPRMTC(89)=IMD(89)+TEMP
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249 IPRMTC(95)=IMD(95)+TEMP
250 IPRMTC(96)=IMD(96)+TEMP
251 IPRMTC(97)=IMD(97)+TEMP
252 IPRMTC(98)=IMD(98)+TEMP
253 IPRMTC(99)=IMD(99)+TEMP
254 IPRMTC(100)=IMD(100)+TEMP

```



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465      605      606      607      608      609      610      611      612      613      614      615      616      617      618      619      620      621      622      623      624      625      626      627      628      629      630      631      632      633      634      635      636      637      638      639      640      641      642      643      644      645      646      647      648      649      650      651      652      653      654      655      656      657      658      659      660      661      662      663      664      665      666      667      668      669      670      671      672      673      674      675      676      677      678      679      680      681      682      683      684      685      686      687      688      689      690      691      692      693      694      695      696      697      698      699      700      701      702      703      704      705      706      707      708      709      710      711      712      713      714      715      716      717      718      719      720      721      722      723      724      725      726      727      728      729      730      731      732      733      734      735      736      737      738      739      740      741      742      743      744      745      746      747      748      749      750      751      752      753      754      755      756      757      758      759      760      761      762      763      764      765      766      767      768      769      770      771      772      773      774      775      776      777      778      779      780      781      782      783      784      785      786      787      788      789      790      791      792      793      794      795      796      797      798      799      800      801      802      803      804      805      806      807      808      809      810      811      812      813      814      815      816      817      818      819      820      821      822      823      824      825      826      827      828      829      830      831      832      833      834      835      836      837      838      839      840      841      842      843      844      845      846      847      848      849      850      851      852      853      854      855      856      857      858      859      860      861      862      863      864      865      866      867      868      869      870      871      872      873      874      875      876      877      878      879      880      881      882      883      884      885      886      887      888      889      890      891      892      893      894      895      896      897      898      899      900      901      902      903      904      905      906      907      908      909      910      911      912      913      914      915      916      917      918      919      920      921      922      923      924      925      926      927      928      929      930      931      932      933      934      935      936      937      938      939      940      941      942      943      944      945      946      947      948      949      950      951      952      953      954      955      956      957      958      959      960      961      962      963      964      965      966      967      968      969      970      971      972      973      974      975      976      977      978      979      980      981      982      983      984      985      986      987      988      989      990      991      992      993      994      995      996      997      998      999      1000

```



```

C --- UFAC=DPCEGSDP/UAB+1
    DO 800 A=1,2
    U(1)=A
    U(2)=A
    U(3)=A
    P(1)=A
    P(2)=A
    P(3)=A
    P(4)=A
    P(5)=A
    P(6)=A
    P(7)=A
    P(8)=A
    P(9)=A
    P(10)=A
    P(11)=A
    P(12)=A
    P(13)=A
    P(14)=A
    P(15)=A
    P(16)=A
    P(17)=A
    P(18)=A
    P(19)=A
    P(20)=A
    P(21)=A
    P(22)=A
    P(23)=A
    P(24)=A
    P(25)=A
    P(26)=A
    P(27)=A
    P(28)=A
    P(29)=A
    P(30)=A
    P(31)=A
    P(32)=A
    P(33)=A
    P(34)=A
    P(35)=A
    P(36)=A
    P(37)=A
    P(38)=A
    P(39)=A
    P(40)=A
    P(41)=A
    P(42)=A
    P(43)=A
    P(44)=A
    P(45)=A
    P(46)=A
    P(47)=A
    P(48)=A
    P(49)=A
    P(50)=A
    P(51)=A
    P(52)=A
    P(53)=A
    P(54)=A
    P(55)=A
    P(56)=A
    P(57)=A
    P(58)=A
    P(59)=A
    P(60)=A
    P(61)=A
    P(62)=A
    P(63)=A
    P(64)=A
    P(65)=A
    P(66)=A
    P(67)=A
    P(68)=A
    P(69)=A
    P(70)=A
    P(71)=A
    P(72)=A
    P(73)=A
    P(74)=A
    P(75)=A
    P(76)=A
    P(77)=A
    P(78)=A
    P(79)=A
    P(80)=A
    P(81)=A
    P(82)=A
    P(83)=A
    P(84)=A
    P(85)=A
    P(86)=A
    P(87)=A
    P(88)=A
    P(89)=A
    P(90)=A
    P(91)=A
    P(92)=A
    P(93)=A
    P(94)=A
    P(95)=A
    P(96)=A
    P(97)=A
    P(98)=A
    P(99)=A
    P(100)=A

C --- CALL STROBE(1)
    GO TO 824

C --- STANDARD GEMIS VERSION
    FLOT=PIE+SI(1)/KI
    DPH=SI(2)/KI
    DP=SI(3)/KI
    DP2=SI(4)/KI
    DP3=SI(5)/KI
    DP4=SI(6)/KI
    DP5=SI(7)/KI
    DP6=SI(8)/KI
    DP7=SI(9)/KI
    DP8=SI(10)/KI
    DP9=SI(11)/KI
    DP10=SI(12)/KI
    DP11=SI(13)/KI
    DP12=SI(14)/KI
    DP13=SI(15)/KI
    DP14=SI(16)/KI
    DP15=SI(17)/KI
    DP16=SI(18)/KI
    DP17=SI(19)/KI
    DP18=SI(20)/KI
    DP19=SI(21)/KI
    DP20=SI(22)/KI
    DP21=SI(23)/KI
    DP22=SI(24)/KI
    DP23=SI(25)/KI
    DP24=SI(26)/KI
    DP25=SI(27)/KI
    DP26=SI(28)/KI
    DP27=SI(29)/KI
    DP28=SI(30)/KI
    DP29=SI(31)/KI
    DP30=SI(32)/KI
    DP31=SI(33)/KI
    DP32=SI(34)/KI
    DP33=SI(35)/KI
    DP34=SI(36)/KI
    DP35=SI(37)/KI
    DP36=SI(38)/KI
    DP37=SI(39)/KI
    DP38=SI(40)/KI
    DP39=SI(41)/KI
    DP40=SI(42)/KI
    DP41=SI(43)/KI
    DP42=SI(44)/KI
    DP43=SI(45)/KI
    DP44=SI(46)/KI
    DP45=SI(47)/KI
    DP46=SI(48)/KI
    DP47=SI(49)/KI
    DP48=SI(50)/KI
    DP49=SI(51)/KI
    DP50=SI(52)/KI
    DP51=SI(53)/KI
    DP52=SI(54)/KI
    DP53=SI(55)/KI
    DP54=SI(56)/KI
    DP55=SI(57)/KI
    DP56=SI(58)/KI
    DP57=SI(59)/KI
    DP58=SI(60)/KI
    DP59=SI(61)/KI
    DP60=SI(62)/KI
    DP61=SI(63)/KI
    DP62=SI(64)/KI
    DP63=SI(65)/KI
    DP64=SI(66)/KI
    DP65=SI(67)/KI
    DP66=SI(68)/KI
    DP67=SI(69)/KI
    DP68=SI(70)/KI
    DP69=SI(71)/KI
    DP70=SI(72)/KI
    DP71=SI(73)/KI
    DP72=SI(74)/KI
    DP73=SI(75)/KI
    DP74=SI(76)/KI
    DP75=SI(77)/KI
    DP76=SI(78)/KI
    DP77=SI(79)/KI
    DP78=SI(80)/KI
    DP79=SI(81)/KI
    DP80=SI(82)/KI
    DP81=SI(83)/KI
    DP82=SI(84)/KI
    DP83=SI(85)/KI
    DP84=SI(86)/KI
    DP85=SI(87)/KI
    DP86=SI(88)/KI
    DP87=SI(89)/KI
    DP88=SI(90)/KI
    DP89=SI(91)/KI
    DP90=SI(92)/KI
    DP91=SI(93)/KI
    DP92=SI(94)/KI
    DP93=SI(95)/KI
    DP94=SI(96)/KI
    DP95=SI(97)/KI
    DP96=SI(98)/KI
    DP97=SI(99)/KI
    DP98=SI(100)/KI

C --- Z FLOW AREAS
    DO 307 A=1,2
    Z(1)=A
    Z(2)=A
    Z(3)=A
    Z(4)=A
    Z(5)=A
    Z(6)=A
    Z(7)=A
    Z(8)=A
    Z(9)=A
    Z(10)=A
    Z(11)=A
    Z(12)=A
    Z(13)=A
    Z(14)=A
    Z(15)=A
    Z(16)=A
    Z(17)=A
    Z(18)=A
    Z(19)=A
    Z(20)=A
    Z(21)=A
    Z(22)=A
    Z(23)=A
    Z(24)=A
    Z(25)=A
    Z(26)=A
    Z(27)=A
    Z(28)=A
    Z(29)=A
    Z(30)=A
    Z(31)=A
    Z(32)=A
    Z(33)=A
    Z(34)=A
    Z(35)=A
    Z(36)=A
    Z(37)=A
    Z(38)=A
    Z(39)=A
    Z(40)=A
    Z(41)=A
    Z(42)=A
    Z(43)=A
    Z(44)=A
    Z(45)=A
    Z(46)=A
    Z(47)=A
    Z(48)=A
    Z(49)=A
    Z(50)=A
    Z(51)=A
    Z(52)=A
    Z(53)=A
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    Z(58)=A
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    Z(64)=A
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    Z(75)=A
    Z(76)=A
    Z(77)=A
    Z(78)=A
    Z(79)=A
    Z(80)=A
    Z(81)=A
    Z(82)=A
    Z(83)=A
    Z(84)=A
    Z(85)=A
    Z(86)=A
    Z(87)=A
    Z(88)=A
    Z(89)=A
    Z(90)=A
    Z(91)=A
    Z(92)=A
    Z(93)=A
    Z(94)=A
    Z(95)=A
    Z(96)=A
    Z(97)=A
    Z(98)=A
    Z(99)=A
    Z(100)=A

C --- W0M2=PIE+SI(1)/KI
    PALFA=0.5*PIE+SI(1)/KI
    DO 225 A=1,2
    W(1)=A
    W(2)=A
    W(3)=A
    W(4)=A
    W(5)=A
    W(6)=A
    W(7)=A
    W(8)=A
    W(9)=A
    W(10)=A
    W(11)=A
    W(12)=A
    W(13)=A
    W(14)=A
    W(15)=A
    W(16)=A
    W(17)=A
    W(18)=A
    W(19)=A
    W(20)=A
    W(21)=A
    W(22)=A
    W(23)=A
    W(24)=A
    W(25)=A
    W(26)=A
    W(27)=A
    W(28)=A
    W(29)=A
    W(30)=A
    W(31)=A
    W(32)=A
    W(33)=A
    W(34)=A
    W(35)=A
    W(36)=A
    W(37)=A
    W(38)=A
    W(39)=A
    W(40)=A
    W(41)=A
    W(42)=A
    W(43)=A
    W(44)=A
    W(45)=A
    W(46)=A
    W(47)=A
    W(48)=A
    W(49)=A
    W(50)=A
    W(51)=A
    W(52)=A
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    W(56)=A
    W(57)=A
    W(58)=A
    W(59)=A
    W(60)=A
    W(61)=A
    W(62)=A
    W(63)=A
    W(64)=A
    W(65)=A
    W(66)=A
    W(67)=A
    W(68)=A
    W(69)=A
    W(70)=A
    W(71)=A
    W(72)=A
    W(73)=A
    W(74)=A
    W(75)=A
    W(76)=A
    W(77)=A
    W(78)=A
    W(79)=A
    W(80)=A
    W(81)=A
    W(82)=A
    W(83)=A
    W(84)=A
    W(85)=A
    W(86)=A
    W(87)=A
    W(88)=A
    W(89)=A
    W(90)=A
    W(91)=A
    W(92)=A
    W(93)=A
    W(94)=A
    W(95)=A
    W(96)=A
    W(97)=A
    W(98)=A
    W(99)=A
    W(100)=A

C --- IF (TEST) 802,801,802
    LAB(1)=801
    LAB(2)=801
    LAB(3)=801
    LAB(4)=801
    LAB(5)=801
    LAB(6)=801
    LAB(7)=801
    LAB(8)=801
    LAB(9)=801
    LAB(10)=801
    LAB(11)=801
    LAB(12)=801
    LAB(13)=801
    LAB(14)=801
    LAB(15)=801
    LAB(16)=801
    LAB(17)=801
    LAB(18)=801
    LAB(19)=801
    LAB(20)=801
    LAB(21)=801
    LAB(22)=801
    LAB(23)=801
    LAB(24)=801
    LAB(25)=801
    LAB(26)=801
    LAB(27)=801
    LAB(28)=801
    LAB(29)=801
    LAB(30)=801
    LAB(31)=801
    LAB(32)=801
    LAB(33)=801
    LAB(34)=801
    LAB(35)=801
    LAB(36)=801
    LAB(37)=801
    LAB(38)=801
    LAB(39)=801
    LAB(40)=801
    LAB(41)=801
    LAB(42)=801
    LAB(43)=801
    LAB(44)=801
    LAB(45)=801
    LAB(46)=801
    LAB(47)=801
    LAB(48)=801
    LAB(49)=801
    LAB(50)=801
    LAB(51)=801
    LAB(52)=801
    LAB(53)=801
    LAB(54)=801
    LAB(55)=801
    LAB(56)=801
    LAB(57)=801
    LAB(58)=801
    LAB(59)=801
    LAB(60)=801
    LAB(61)=801
    LAB(62)=801
    LAB(63)=801
    LAB(64)=801
    LAB(65)=801
    LAB(66)=801
    LAB(67)=801
    LAB(68)=801
    LAB(69)=801
    LAB(70)=801
    LAB(71)=801
    LAB(72)=801
    LAB(73)=801
    LAB(74)=801
    LAB(75)=801
    LAB(76)=801
    LAB(77)=801
    LAB(78)=801
    LAB(79)=801
    LAB(80)=801
    LAB(81)=801
    LAB(82)=801
    LAB(83)=801
    LAB(84)=801
    LAB(85)=801
    LAB(86)=801
    LAB(87)=801
    LAB(88)=801
    LAB(89)=801
    LAB(90)=801
    LAB(91)=801
    LAB(92)=801
    LAB(93)=801
    LAB(94)=801
    LAB(95)=801
    LAB(96)=801
    LAB(97)=801
    LAB(98)=801
    LAB(99)=801
    LAB(100)=801

C --- TEST 3
    LAB(1)=802
    LAB(2)=802
    LAB(3)=802
    LAB(4)=802
    LAB(5)=802
    LAB(6)=802
    LAB(7)=802
    LAB(8)=802
    LAB(9)=802
    LAB(10)=802
    LAB(11)=802
    LAB(12)=802
    LAB(13)=802
    LAB(14)=802
    LAB(15)=802
    LAB(16)=802
    LAB(17)=802
    LAB(18)=802
    LAB(19)=802
    LAB(20)=802
    LAB(21)=802
    LAB(22)=802
    LAB(23)=802
    LAB(24)=802
    LAB(25)=802
    LAB(26)=802
    LAB(27)=802
    LAB(28)=802
    LAB(29)=802
    LAB(30)=802
    LAB(31)=802
    LAB(32)=802
    LAB(33)=802
    LAB(34)=802
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    LAB(36)=802
    LAB(37)=802
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    LAB(73)=802
    LAB(74)=802
    LAB(75)=802
    LAB(76)=802
    LAB(77)=802
    LAB(78)=802
    LAB(79)=802
    LAB(80)=802
    LAB(81)=802
    LAB(82)=802
    LAB(83)=802
    LAB(84)=802
    LAB(85)=802
    LAB(86)=802
    LAB(87)=802
    LAB(88)=802
    LAB(89)=802
    LAB(90)=802
    LAB(91)=802
    LAB(92)=802
    LAB(93)=802
    LAB(94)=802
    LAB(95)=802
    LAB(96)=802
    LAB(97)=802
    LAB(98)=802
    LAB(99)=802
    LAB(100)=802

```



```

1005 C
1 SM ER=12.7M BR=1PEL1.3.4M PSI=ELL.3.4M PSIE=ELL.3.
2 SM ER=12.7M BR=1PEL1.3.4M PSI=ELL.3.4M PSIE=ELL.3.
3 SM ER=12.7M BR=1PEL1.3.4M PSI=ELL.3.4M PSIE=ELL.3.

1020 C
URAB=0
DO 1020 J=1,MF
AJED(J)=0
FLUX(J)=0
PFLUX=0
DO 1021 I=1,AMP
MUT(I)=U(I)
PFLUX=PFLUX+MUT(I)*PFLUX(I)
DO 1021 I=1,MF
IF I=SO,VE(J),EQ,01 GO TO 1021
FLUX(J)=FLUX(J)+OMB(I)*(F(J)-F(J+1))
1021 CONTINUE
WFLUX=PFLUX+PFLUX
DO 1022 J=1,MF
FLUX(J)=.5*PFLUX(J)
C
URAB=URAB
MUT=VE(I).5/DIR(I)*(OMB(I)+T(MP3))
DO 1023 J=1,MF
IF I=SO,VE(J),EQ,01 GO TO 1023
OPI(J)=LUT(J)*PEI*(F(J),I)
COM1(MP3)=PEI*(F(J)+F(J+1)-F(J),MP3)
1023 CONTINUE
PFLUX=URAB*PSIE*(MP3)+OPI(MP3)
FLUX(J)=PFLUX(J)+PSIE*2*OPI(J)
FLUX(M)=PFLUX(M)+PSIE*FID*LABPS(I)
C
WRITE (6,1031) BR,URAB,FLUX(J),J,MF
1031 FORMAT (M 20,1PEL1.3.7M OFLUX=ELL.3.4M FLUX(J)=TELL.3/42F,
1 CEL1.3)
C
IF(I=NE,1) GO TO 1024
MUT(I)=MUT+PFLUX
DO 1024 J=1,MF
IF I=SO,VE(J),EQ,01 GO TO 1024
AJED(J)=AJED(J)+MUT*FDFE(J)
1024 CONTINUE
COM1(MP3)=MUT*TAUO*(MUT(I)+1,MF)
1020 PFLUX=1.3M ER=12.7M TAUEB=1PEL1.3.4M AJED(J)=TELL.3/34F,
1 CEL1.3)
C
1024 IF(URAB=1) GO TO 1026
TAUEB=TAUEB+MUT*PFLUX
DO 1025 J=1,MF
IF I=SO,VE(J),EQ,01 GO TO 1027
AJED(J)=AJED(J)+MUT*FDFE(J)
1027 CONTINUE
MUT=URAB
1020 PFLUX=1.3M ER=12.7M TAUEB=1PEL1.3.4M AJED(J)=TELL.3/34F,
1020 CONTINUE
C
CHAPTER 1028 ER=12.7M TAUEB=1PEL1.3.4M AJED(J)=TELL.3/34F,
1028 CONTINUE
C
IF(I=PRET,EG,1) GO TO 110
C
LAB11=MBL1.7'S
LAB12=GM
D1W=1
DO 1095 I=1,AMP
DUT(I)=PFLUX(I)*D1W
MUT(I)=MUT(I)
DO 1095 I=1,MF
MUT(I)=DUT(I)
1095 CONTINUE
LAB11=D1W
LAB12=GM
SUB=0
D1W=1
IF(URAB,EG,3) D1W=J11+1.E-30

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```

1      FUNCTION TAB (I, II, IV, RTAB)
2      DIMENSION X(1), Y(1)
3      TAB=0
4      GO TO 5
5      RETURN
6      IF (X(1)-GT, XX(2)) F--F
7      DO 10 J=1, RTAB
8      IF (F<X(1)-X(J)) GO TO 20
9      CONTINUE
10     IF (I.NE.1) GO TO 30
11     I=1
12     DEL=XX(1)-XX(J)
13     IF (DEL<EQ, 0) GO TO 50
14     TAB=(Y(I)+I*DEL**2)/(X(I)-XX(J))/DEL
15     RETURN
16     GO TO 1
17     WRITE (6,60) I, I, J
18     CALL EXIT
19     COMPA 1' *** ERROR IN SUBROUTINE TAB ***
20     END

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```

CARD NR. SEVERITY DETAILS DIAGNOSIS OF PROBLEM
7 I XI ARRAY REFERENCE OUTSIDE DIMENSION BOUNDS.

FUNCTION VISCO 73774 OPT-3 TRACE
 FUNCTION VISCO(1)
 VISCO(1)=0000-30(1/256,5)000-76
 RETURN
 END

FTR 4.0-430
 08/24/78 14.02.39
 PAGE 1
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```

195 506-7506(LM,MP2)=-2506(LM,MP1)
196 570-7506(LM,MP2)=-2506(LM,MP1)
197 501(MP2)=EMUL(MP2)0CEM(MP2)
198 502(MP2)=EMUL(MP2)0CEM(MP2)
199 503(MP2)=EMUL(MP2)0CEM(MP2)
200 504(MP2)=EMUL(MP2)0CEM(MP2)
201 505(MP2)=EMUL(MP2)0CEM(MP2)
202 506(MP2)=EMUL(MP2)0CEM(MP2)
203 507(MP2)=EMUL(MP2)0CEM(MP2)
204 508(MP2)=EMUL(MP2)0CEM(MP2)
205 509(MP2)=EMUL(MP2)0CEM(MP2)
206 510(MP2)=EMUL(MP2)0CEM(MP2)
207 511(MP2)=EMUL(MP2)0CEM(MP2)
208 512(MP2)=EMUL(MP2)0CEM(MP2)
209 513(MP2)=EMUL(MP2)0CEM(MP2)
210 514(MP2)=EMUL(MP2)0CEM(MP2)
211 515(MP2)=EMUL(MP2)0CEM(MP2)
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214 518(MP2)=EMUL(MP2)0CEM(MP2)
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217 521(MP2)=EMUL(MP2)0CEM(MP2)
218 522(MP2)=EMUL(MP2)0CEM(MP2)
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222 526(MP2)=EMUL(MP2)0CEM(MP2)
223 527(MP2)=EMUL(MP2)0CEM(MP2)
224 528(MP2)=EMUL(MP2)0CEM(MP2)
225 529(MP2)=EMUL(MP2)0CEM(MP2)
230 100 PFORMAT(14,16,42,10)11E11.3/(9K,11E11.3)

```

END 40 220

CARD NO. SEVERITY DETAILS DIAGNOSIS OF PROBLEM

187 I AM IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.

206 I AM IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.

AM IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.

AM IF STATEMENT MAY BE MORE EFFICIENT THAN A 2 OR 3 BRANCH COMPUTED GO TO STATEMENT.


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195 0417445(3)F04(MH)FCR(MW)S)00DK2
196 0417446(1)F04(MH)FCR(MW)S)00DK2
197 0417447(1)F04(MH)FCR(MW)S)00DK2
198 0417448(1)F04(MH)FCR(MW)S)00DK2
199 0417449(1)F04(MH)FCR(MW)S)00DK2
200 0417450(1)F04(MH)FCR(MW)S)00DK2
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210 0417460(1)F04(MH)FCR(MW)S)00DK2
211 0417461(1)F04(MH)FCR(MW)S)00DK2
212 0417462(1)F04(MH)FCR(MW)S)00DK2
213 0417463(1)F04(MH)FCR(MW)S)00DK2
214 0417464(1)F04(MH)FCR(MW)S)00DK2
215 0417465(1)F04(MH)FCR(MW)S)00DK2
216 0417466(1)F04(MH)FCR(MW)S)00DK2
217 0417467(1)F04(MH)FCR(MW)S)00DK2
218 0417468(1)F04(MH)FCR(MW)S)00DK2
219 0417469(1)F04(MH)FCR(MW)S)00DK2
220 0417470(1)F04(MH)FCR(MW)S)00DK2
221 0417471(1)F04(MH)FCR(MW)S)00DK2
222 0417472(1)F04(MH)FCR(MW)S)00DK2
223 0417473(1)F04(MH)FCR(MW)S)00DK2
224 0417474(1)F04(MH)FCR(MW)S)00DK2
225 0417475(1)F04(MH)FCR(MW)S)00DK2
226 0417476(1)F04(MH)FCR(MW)S)00DK2
227 0417477(1)F04(MH)FCR(MW)S)00DK2
228 0417478(1)F04(MH)FCR(MW)S)00DK2
229 0417479(1)F04(MH)FCR(MW)S)00DK2
230 0417480(1)F04(MH)FCR(MW)S)00DK2

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235      MPROM=PELOBM(I172.
          DC 3000, 1.0, 1.950
          IF (LARG1(1,1)) GOTO 236
          DELTA=ABS(1.0-0.950)/ABS(1.0-0.950)
          CONTINUE
          XDI=XDI+DHI
          IF (XDI-51.50) GOTO 240
          IF (XDI-51.50) GOTO 240
          LAB(1)=MAUSZ
          LAB(2)=24
          WRITE (6,900) LAB(1),XDI,STEP,COUNT
          900  WRITE (6,900) LAB(1),XDI,STEP,COUNT
          910  FORMAT (140,6,2,15,11.3/102,11E11.3)
          920  STOP
C
240      C
          C 010 PROM=PELOBM(I170KI
          C 010 SOLVE FOR DOWNSTREAM SPECIE VALUES
          DO 730 J=1,N3
            P1=PMO*(U(J)+2.0)*TSOR(I3)*SDBR(I3)*DME(I3)/P1BDM+2.0*950M(I3)/
            P1(J)
          730  CONTINUE
          XDI=XDI
          IF (XDI-51.50) GOTO 700
          PROM=PELOBM(I170KI
          740  FORMAT (140,6,2,15,11.3/102,11E11.3)
          750  STOP
          760  WRITE (6,910) XDI,STEP,COUNT
          910  FORMAT (140,6,2,15,11.3/102,11E11.3)
          770  CONTINUE
          RETURN
          END

```



```

80 IF (I=EQ) T=AMN(I)*I/M
   IF (I=1) T=AMN(I)*I/M
   GO TO 25
20 ARG=AL(I)*T+AL(2)*T+AL(3)*T+AL(4)*T+AL(5)*T+AL(6)*T+AL(7)*T+AL(8)*T+AL(9)*T+AL(10)*T
   RETURN
25 CASP=ARG*1.98741647M/I
   *****
   *****
   *****
85 C *****
   *****
   *****
   *****
90 IF (I=EQ) T=AMN(I)*I/M
   IF (I=1) T=AMN(I)*I/M
   ARG=AL(I)*T+AL(2)*T+AL(3)*T+AL(4)*T+AL(5)*T+AL(6)*T+AL(7)*T+AL(8)*T+AL(9)*T+AL(10)*T
   GO TO 35
30 ARG=AL(I)*T+AL(2)*T+AL(3)*T+AL(4)*T+AL(5)*T+AL(6)*T+AL(7)*T+AL(8)*T+AL(9)*T+AL(10)*T
   CASP=ARG*1.98741647M/I
   *****
   *****
   *****
95 IF (I=EQ) T=AMN(I)*I/M
   IF (I=1) T=AMN(I)*I/M
   ARG=AL(I)*T+AL(2)*T+AL(3)*T+AL(4)*T+AL(5)*T+AL(6)*T+AL(7)*T+AL(8)*T+AL(9)*T+AL(10)*T
   CASP=ARG*1.98741647M/I
   *****
   *****
   *****
100 C *****
   *****
   *****
   *****
105 *****
   *****
   *****
   *****
110 *****
   *****
   *****
   *****

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```

1106 R(1)=V(1)/CSALFA
      V(1)=ABS(V(1)/CSALFA)
      GO TO 1107
1107
1108 R(2)=V(2)/CSALFA
      V(2)=ABS(V(2)/CSALFA)
      GO TO 1109
1109
1110 R(3)=V(3)/CSALFA
      V(3)=ABS(V(3)/CSALFA)
      GO TO 1111
1111 V(4)=V(4)/CSALFA
      V(5)=V(5)/CSALFA
      V(6)=V(6)/CSALFA
      V(7)=V(7)/CSALFA
      V(8)=V(8)/CSALFA
      RETURN
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00/24/78 14.42.35

FTM 4.0-439

SUBROUTINE PLOTS 73/74 OPT=0 TRACE

```

00  59 AICI=BLANK
    60 CONTINUE
    70  IF (I) GO TO (A111-1)
    80  WRITE(6,104) (A11),I-1,11)
    90  RETURN
    100 FORMAT(1M 'VALUES',5L,10I1L,46,A6)
    110  CONTINUE
    120  CONTINUE
    130  CONTINUE
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    990  CONTINUE
    1000 CONTINUE
  
```


APPENDIX G

LISTING OF FUEL INSTRUCTION MODEL

PAGE 1

06-724778 14-28-13

FTM 4.6-435

OPT-6 TRACE

PROGRAM INJECTI 73774

```

1  PROGRAM INJECTI (INPUT,OUTPUT,TAPEO,IMPDT,ARF6I=OUTPUT)
2  C
3  C
4  C
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3  COMMON /FERRC/ I, J, K, L, M, N, O, P, Q, R, S, T, U, V, W, X, Y, Z
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5776/79 14.25.13

FTM 4.55436

73/74 CRT-5 TRACE

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257 NCHANS1527, N
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1 SUBROUTINE ENAP (LIMIT=1)
2 COMPLETE PARADIGMATION.
3 PROGRAMMED 1-22-71 BY W. TAYL.
4 CCCCC THIS ROUTINE IS THE SAME AS IM PROGRAM IMPACT EXCEPT
5 FOR THE CARD CHANGE MARKER CASE
6
7 DATE, WFE
8 WFE --- FULL PROPERTY FOOTPRINTS ARE ONLY FOR 304 AND 485/APP 000
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77 COMMON /IMBUS/ IMBUS(20), DEC(2), CAR(20), TITLE(80)

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155 VF2=SQRT(DUM1**2+DUM2**2)
    GO TO 120
160 C 110 DELT=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
165 C 120 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
170 C 130 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
175 C 140 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
180 C 150 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
185 C 160 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
190 C 170 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
195 C 180 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
200 C 190 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
205 C 200 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
210 C 210 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
215 C 220 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
220 C 230 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
225 C 240 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140
230 C 250 DELTA=DELTA*DELTA
    IF (ABS(DELTA-DELTA)) < .01) GO TO 140
    IF (ABS(DELTA-DELTA)) > .01) GO TO 140

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09/24/78 16:29:33

FTN 4-0-430

73/74 OPT=0 TRACE

SUBROUTINE EVAP

```

REC(1)=WF2
IF(SAVE(1)=TF(1))
  IF(J1=1)
    OPT=0)=-0.11
  ELSE
    OPT=0.1)=-0.11
  ENDIF
ENDIF
RETURN

```

EVA 236
EVA 235
EVA 234
EVA 233
EVA 232
EVA 231
EVA 230
EVA 229
EVA 228
EVA 227
EVA 226
EVA 225
EVA 224

235

240

245

```

C 210 FORMAT (10H0000 EVAP FATLJBF.415.SF.74HROBLET15)
220 FORMAT (4E15.5)
230 END

```



```

FUNCTION VAPCPF 73/74 OPT=O TRACE          08/24/78 14.28.13      PAGE 1
1      C
      FUNCTION VAPCPF (I)
      Y=TEMPER (I)* VAPCPF= CP FOR JPA JPS/JPS FUEL VAPED
      COMBIN /INPUT/ MSEC (REG-CAP-PH-TITLE)
      IF (FUEL (IC-2) GO TO 15
      CAP=1.66*0.00563*F
      10 VAPCF=0.04*0.002330*F
      20 RETURN
      END

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1      C      FUNCTION VAPRUF (I)
2      C      IN TEMPER (BT), VAPRUF= MJ FOR JPS, JPS/JPP FBEL VAPOR
3      C      COMPUTED FROM JPS, JPP, JPP/JPP FBEL VAPOR
4      C      VAPRUF=0.0E+00, 0.0E+00
5      C      GO TO 20
6      C      18 VAPRUF=248.E-9+0.31E-9*Y
7      C      20 RETURN
8      C      END
9      C      11
10     C      12
11     C      13
12     C      14
13     C      15
14     C      16
15     C      17
16     C      18
17     C      19
18     C      20

```


0872470 14.29.13

FTN 4.6430

FUNCTION SECT 7374 OPT=0 TRACE

PRDP 32
PRDP 33
PRDP 34
PRDP 35
PRDP 36
PRDP 37

FUNCTION SECT 11250000
SCM - SC RESUME AT 02 DES F
L. 11250000. 11250000. 02 DES F
L. 11250000. 11250000. 02 DES F
RETURN
END

1 C
S

```

1      FUNCTION SCRIPT (E)
      E- REACTION EVAPORATED, S200- 50 AT 60 DEG F
      SCRIPT - 30 051000 01 00 210 0000 0000 0000 0000
      DATA - 00000000 00000000 00000000 00000000
      S200- 5000 0000 0000 0000 0000 0000 0000 0000
      IF (RFDL.ME.4) S660-56000
      SCRIPT-1.0767(1.07675600-1.1001.--070E101.)
      END
10

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PROP
PROP
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08/26/74 14.28.13

FTM 4.0-439

7374 OPT-0 TRACE

FUNCTION VAPOR

FUNCTION VAPOR (VAP)
 VAPOR (VAP) = STABLE PRESSURE
 VAPOR DIFFUSIVITY = 0.001
 VAPOR = 2.003/101101.1101-0.1.0752-0.1-9.9152-0.1
 RETURN
 END

PROP
 PROP
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 PROP
 PROP

48
 50
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0872472 14.78.13

FTM 4.00030

FUNCTION VAPLRF 73/74 OPT=0 TRACE

PPDP	65
PPDP	67
PPDP	68
PPDP	70
PPDP	72
PPDP	73
PPDP	74
PPDP	75
PPDP	77

FUNCTION VAPLRF (1)
T. TEMPER IN VAPLRF. LATENT HEAT TR RECOMBINATION FOR JPA, BTU/LB

IF (KXUEL.EQ.2) GO TO 10
COMMON /INPUT/ KXUEL, WGT, SLP, PP4, TITLE(4)

VAPLRF=1.092.88) VAPLRF=13.20*(1092.33-7100.30
20 VAPLRF=0.

10 IF (T.CT.1210.45) VAPLRF=10.75*(1210.45-1100.41
20 RETURN
END

C

1

5

10

FUNCTION CPFC 73/74 CPT-0 TRACE 08/28/78 16.26.13 PAGE 1

1 E FUNCTION CPFC IT-5CR603
V. TIME (R) CPFC, C FOR LTOUO J%, JPS/JPS
SR60-56 RESIDE AT 60 DEC F FOR J%
CPFC (-1810-0004-901)/50RT(S-50)
END

PROF
PROF
PROF
PROF
PROF
PROF

78
79
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81
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83

FTM 4.6-439

SUBROUTINE SCALES 73/74 OPT=0 TRACE

```

1  SUBROUTINE SCALES (MAX*MIN, LENGTH, STPP, START)
2  REAL MAX*MIN, LENGTH
3  DIMENSION I(1)
4  DIMENSION I(1)
5  I(1) = 0
6  I(1) = 1
7  I(1) = 2
8  I(1) = 3
9  I(1) = 4
10 I(1) = 5
11 I(1) = 6
12 I(1) = 7
13 I(1) = 8
14 I(1) = 9
15 I(1) = 10
16 I(1) = 11
17 I(1) = 12
18 I(1) = 13
19 I(1) = 14
20 I(1) = 15
21 I(1) = 16
22 I(1) = 17
23 I(1) = 18
24 I(1) = 19
25 I(1) = 20
30 CONTINUE
40 FORMAT (//)
END

```

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