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# HYPERSONIC STATIC STABILITY INVESTIGATION OF A TOMAHAWK 20-PERCENT-SCALE MODEL

Glenn H. Merz  
ARO, Inc.

August 1966

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## FOREWORD

The work reported herein was done at the request of NASA, Goddard Space Flight Center, for Fairchild-Hiller Corporation under System 921E.

The results of tests presented were obtained by ARO, Inc. (a subsidiary of Sverdrup & Parcel and Associates, Inc.), contract operator of the Arnold Engineering Development Center (AEDC), Air Force Systems Command (AFSC), Arnold Air Force Station, Tennessee, under Contract AF40(600)-1200. The test was conducted from May 27 to June 2, 1966, under ARO Project No. VB1686, and the manuscript was submitted for publication on July 14, 1966.

The author wishes to thank L. L. Trimmer, ARO, Inc., for his technical assistance during these tests.

This technical report has been reviewed and is approved.

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**ABSTRACT**

An experimental investigation was conducted to obtain detailed static stability, induced roll and yaw, and roll forcing information on several configurations of a Tomahawk 20-percent-scale model. Tests were conducted at nominal Mach numbers of 6 and 8, free-stream Reynolds numbers of  $2.5 \times 10^6$  and  $5.0 \times 10^6$  per foot, angles of attack from -2 to 14 deg, and roll angles from 0 to 90 deg. Selected results are presented to illustrate types and quality of data obtained.

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## NOMENCLATURE

$C_l$	Rolling-moment coefficient, rolling moment/ $q_\infty Sd$
$C_m$	Pitching-moment coefficient, pitching moment*/ $q_\infty Sd$
$C_N$	Normal-force coefficient, normal force*/ $q_\infty S$
$C_n$	Yawing-moment coefficient, yawing moment*/ $q_\infty Sd$
$C_Y$	Side-force coefficient, side force*/ $q_\infty S$
$\Delta C_l$	Differential rolling-moment coefficient, $C_{l_\alpha} - C_{l_{\alpha=0}}$
$\Delta C_n$	Differential yawing-moment coefficient, $C_{n_\alpha} - C_{n_{\alpha=0}}$
$\Delta C_Y$	Differential side-force coefficient, $C_{Y_\alpha} - C_{Y_{\alpha=0}}$
$d$	Model centerbody diameter, 1.80 in.
$M_\infty$	Free-stream Mach number
$P_0$	Tunnel stagnation pressure, psia
$q_\infty$	Free-stream dynamic pressure, psia
$Re_\infty$	Free-stream Reynolds number, $ft^{-1}$
$S$	Model centerbody cross-sectional area, 2.545 in. <sup>2</sup>
$T_0$	Tunnel stagnation temperature, °R
$X_{cp}$	Center of pressure, $10.539 + C_m/C_N$ , diameter from base
$\alpha$	Angle of attack, deg
$\phi$	Roll angle, deg

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\*Measured in ballistic axis

## SECTION I INTRODUCTION

The purpose of these tests was to obtain detailed static stability, induced roll and yaw, and roll forcing information on a Tomahawk rocket vehicle 20-percent-scale model at Mach 6 and 8. Model centerbody extensions were used to obtain configurations of three different lengths. The medium length model was tested with three nose and three fin configurations. Induced roll and yaw tests were conducted only at Mach 6.

Tests were conducted in the von Kármán Gas Dynamics Facility (VKF) 50-in. hypersonic tunnel (Gas Dynamic Wind Tunnel, Hypersonic (B)) at Mach numbers of 6 and 8, free-stream unit Reynolds numbers of  $2.5 \times 10^6$  and  $5.0 \times 10^6$ , angles of attack from -2 to 14 deg, and roll angles from 0 to 90 deg.

## SECTION II APPARATUS

### 2.1 MODELS

Model photographs and details are shown in Figs. 1 and 2. The various configurations consisted of three nose sections, two ogive and one conical; three centerbody extensions; and three sets of flaps, two without cant and one with a 1-deg positive cant.

### 2.2 WIND TUNNEL

Tunnel B is a 50-in. -diam continuous flow, closed-circuit, variable density wind tunnel equipped with interchangeable Mach 6 and 8 axisymmetric, contoured nozzles. It operates at nominal Mach numbers of 6 and 8 at stagnation pressures from 20 to 280 psia and from 50 to 900 psia, respectively, at temperatures up to 1350°R.

Figure 3 shows the tunnel and its associated equipment. As shown in this figure, the model may be injected into the tunnel for a test run and then retracted from the test section for model changes without interrupting tunnel airflow.

## 2.3 INSTRUMENTATION

A six-component, moment-type, strain-gage balance was used to measure model forces and moments. Prior to testing, a static calibration was performed using combined loads of the magnitude anticipated during this test. The expected deviations, based on this calibration, are summarized below:

<u>Component</u>	<u>Deviation</u>
Normal Force	±0.15 lb
Pitching Moment	±0.50 in. -lb
Side Force	±0.05 lb
Yawing Moment	±0.15 in. -lb
Rolling Moment	±0.05 in. -lb

The data presented in Section IV were obtained by rolling the model on the balance to obtain ballistic axes data directly; thus, the above-mentioned deviations apply directly to the data.

### SECTION III TEST CONDITIONS

Conditions used during the tests are summarized as follows:

<u>Configuration</u>	<u>M<sub>∞</sub></u>	<u>P<sub>O</sub>, psia</u>	<u>T<sub>O</sub>, °R</u>	<u>Re<sub>∞</sub> × 10<sup>-6</sup></u>
N <sub>1</sub> E <sub>2</sub> B <sub>1</sub> F <sub>1</sub>	6.06	280	860	5.0
All	6.03	150	860	2.5
All	7.99	545	1310	2.5

### SECTION IV RESULTS AND DISCUSSION

Selected data at Mach 6, zero roll,  $Re_{\infty}/ft = 2.5 \times 10^6$  are presented in Fig. 4 to illustrate types of data obtained for various length models. As would be expected, the center of pressure moved forward on the model as the length was increased. Typical data for the basic configuration are presented in Fig. 5 to illustrate the effect of Mach number on the stability characteristics.

Figure 6 is a summary of the Mach 6 lateral stability data at various roll angles. Canting the fins 1 deg produced a significant change in the

rolling-moment data over the entire angle-of-attack range. However, it did not appreciably alter the side-force and yawing-moment data at the lower angles of attack.

The differential lateral stability characteristics are presented in Fig. 7. These coefficients were computed by subtracting the zero angle-of-attack data from the data obtained at various angles of attack. The canted fins had little effect on the incremental forces and moments produced by changing angle of attack.

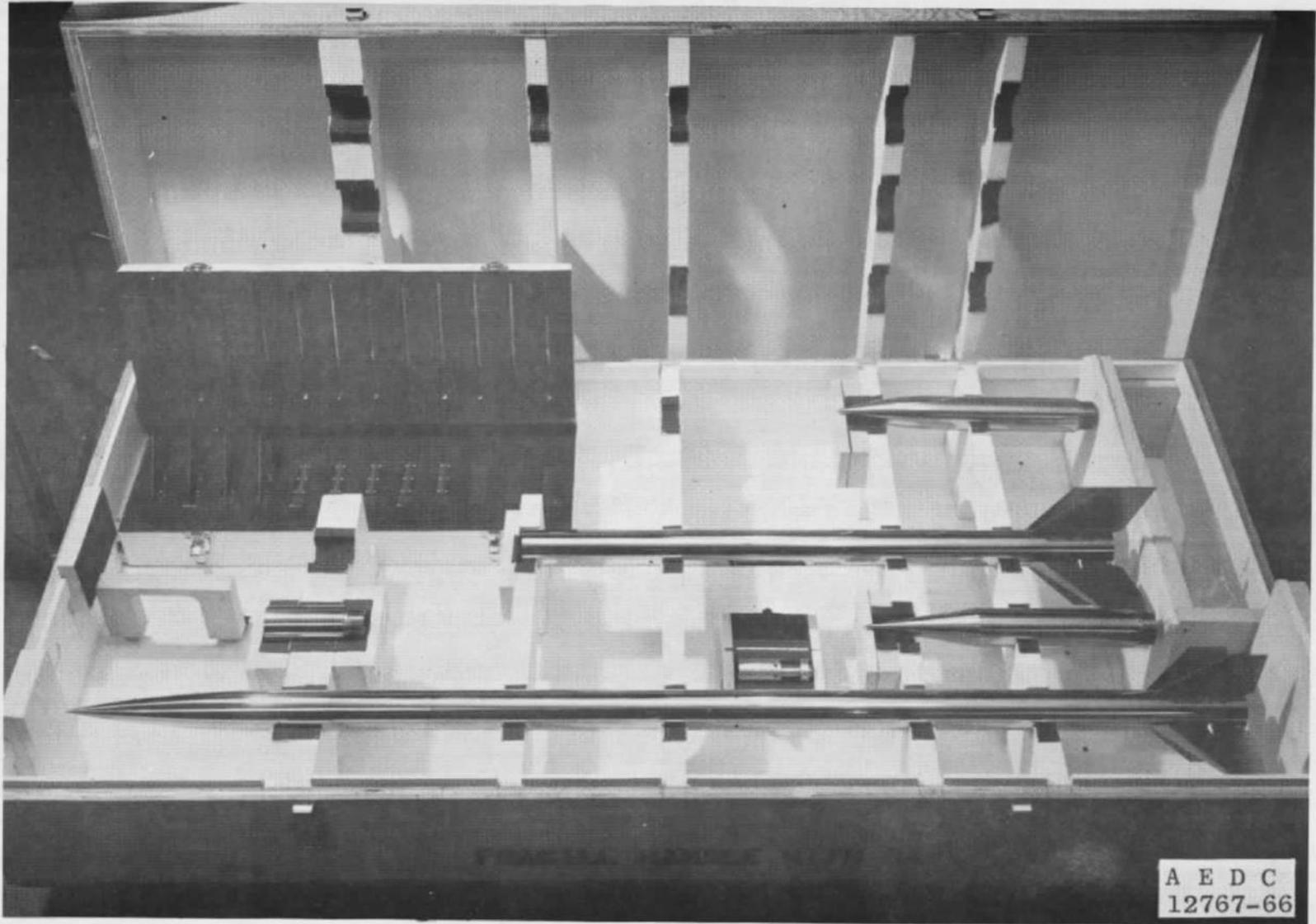
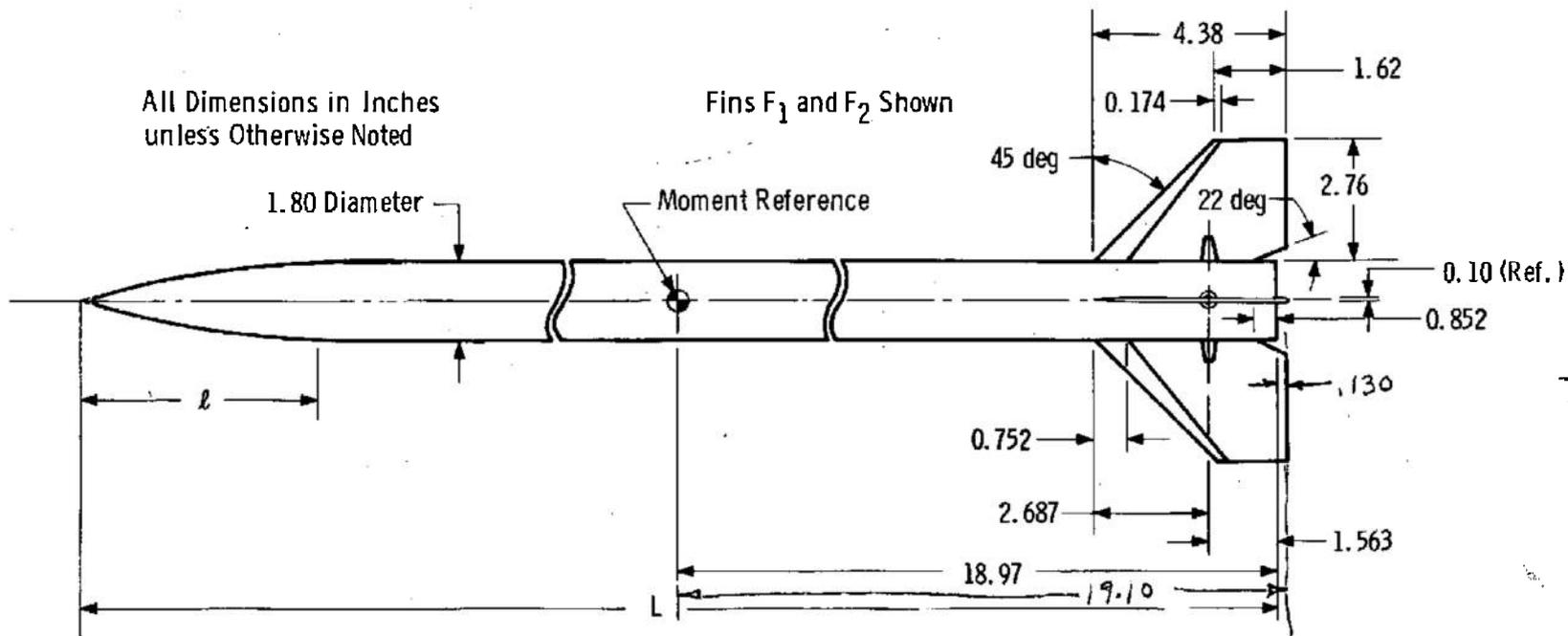


Fig. 1 Model Photograph

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7.32 span

Nose	Type	$l$
N <sub>1</sub>	Ogive	5.4
N <sub>2</sub>	Conical	5.4
N <sub>3</sub>	Ogive	9.0

Extension	L
E <sub>1</sub>	40.8
E <sub>2</sub>	44.4
E <sub>3</sub>	48.0

Fin	Type
F <sub>1</sub>	0-deg Cant
F <sub>2</sub>	1-deg Positive Cant
F <sub>3</sub>	0-deg Cant

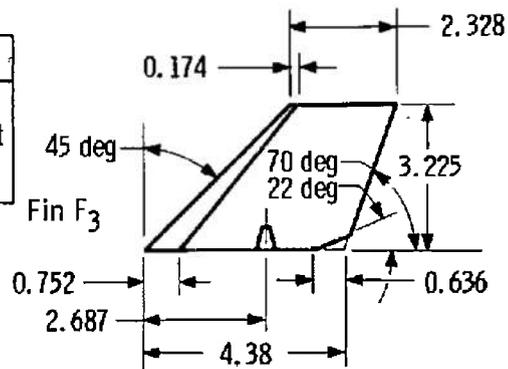
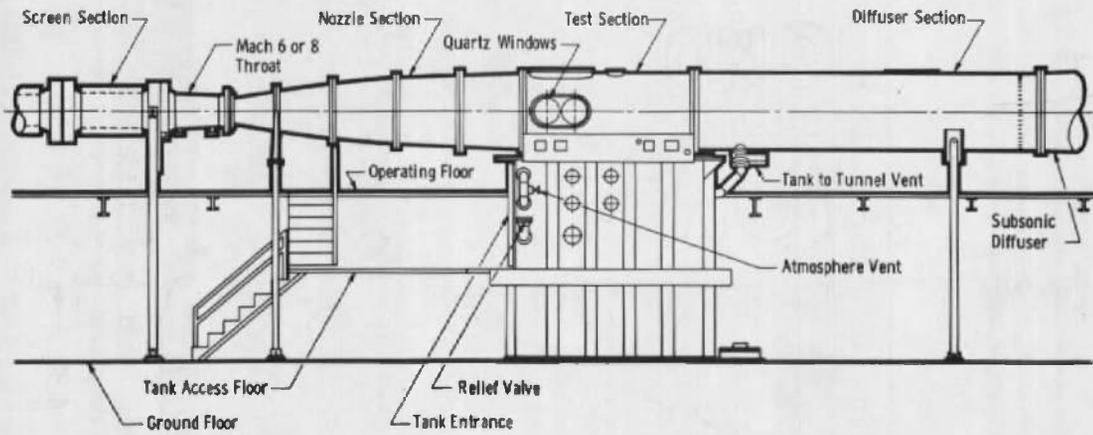
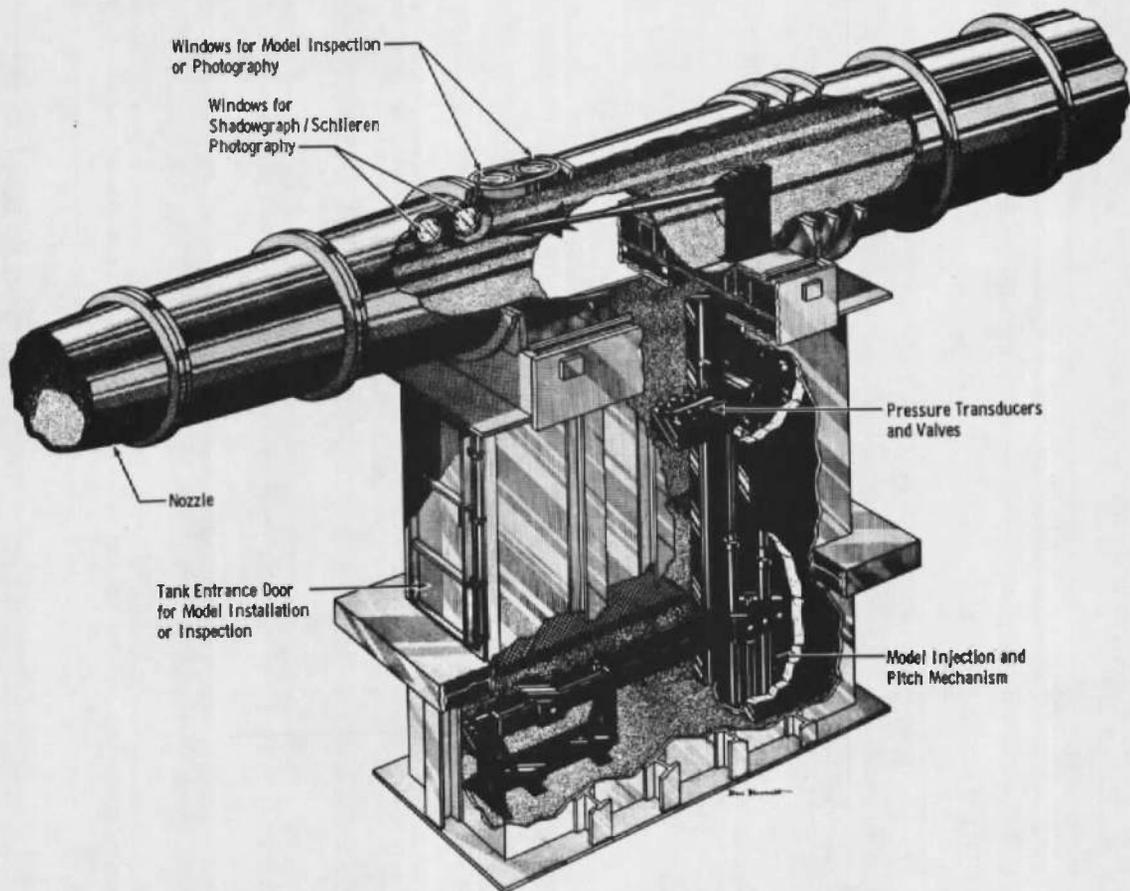


Fig. 2 Model Details



**Tunnel Assembly**



**Tunnel Test Section**

**Fig. 3 Tunnel B**

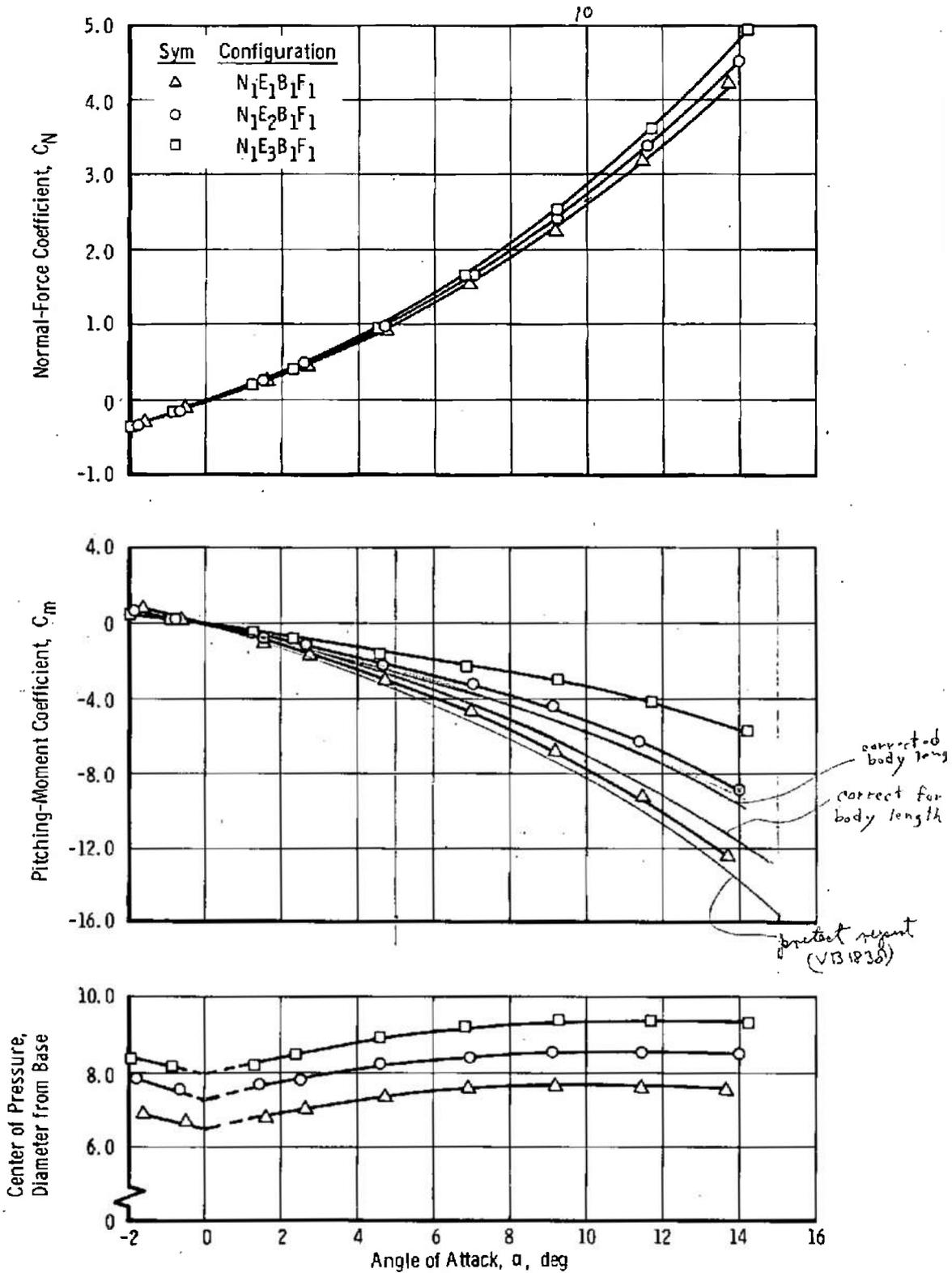


Fig. 4 Effect of Model Length on Longitudinal Stability Characteristics at Mach 6, Zero Roll,  $Re_{\infty}/ft = 2.5 \times 10^6$

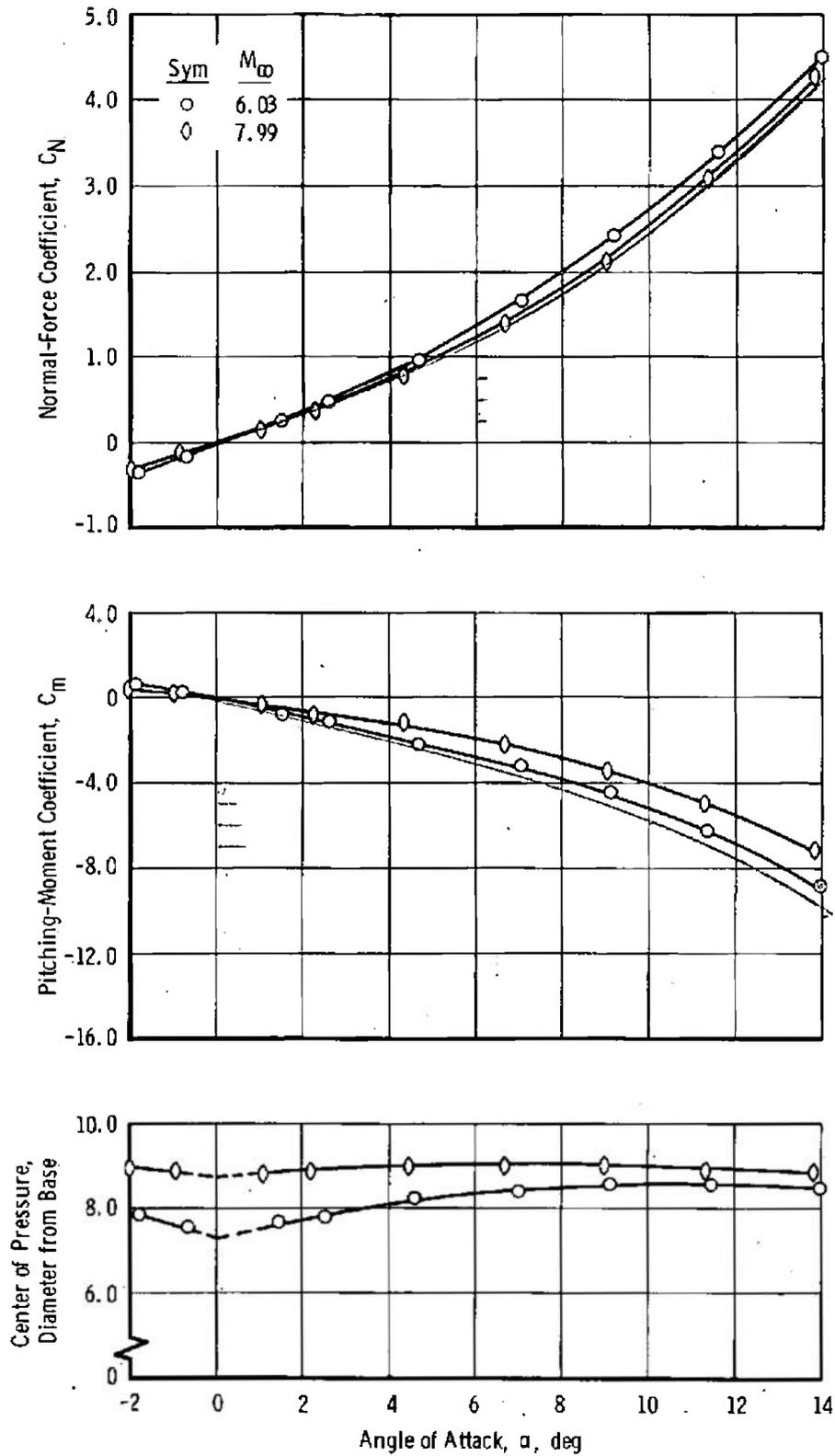


Fig. 5 Effect of Mach Number on Longitudinal Stability Characteristics of Configuration N1E2B1F1, Zero Roll,  $Re_\infty/ft = 2.5 \times 10^6$

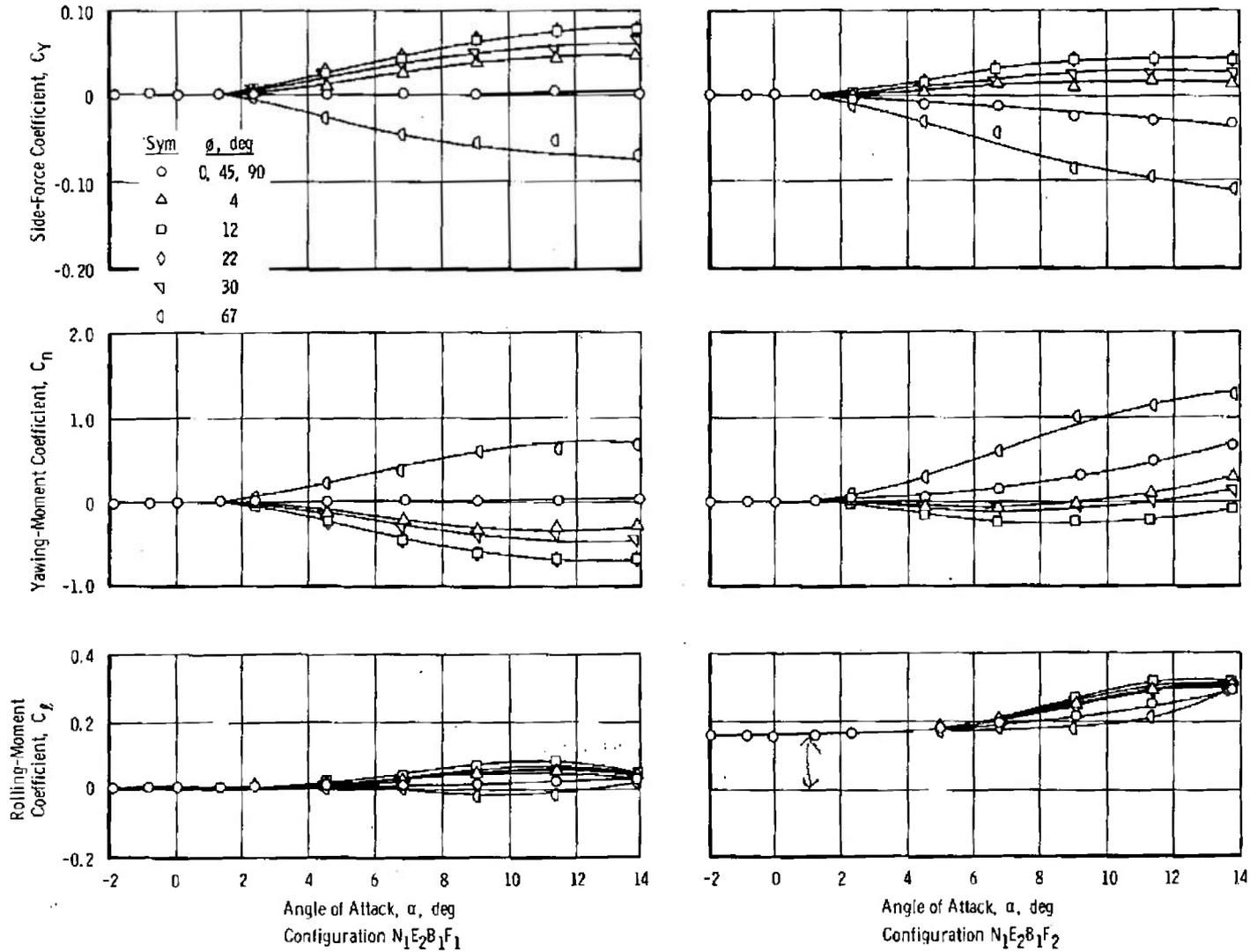


Fig. 6 Effect of Roll Angle on Lateral Stability Characteristics of Configurations  $N_1E_2B_1F_1$  and  $N_1E_2B_1F_2$  at Mach 6,  $Re_\infty/ft = 2.5 \times 10^6$

.005

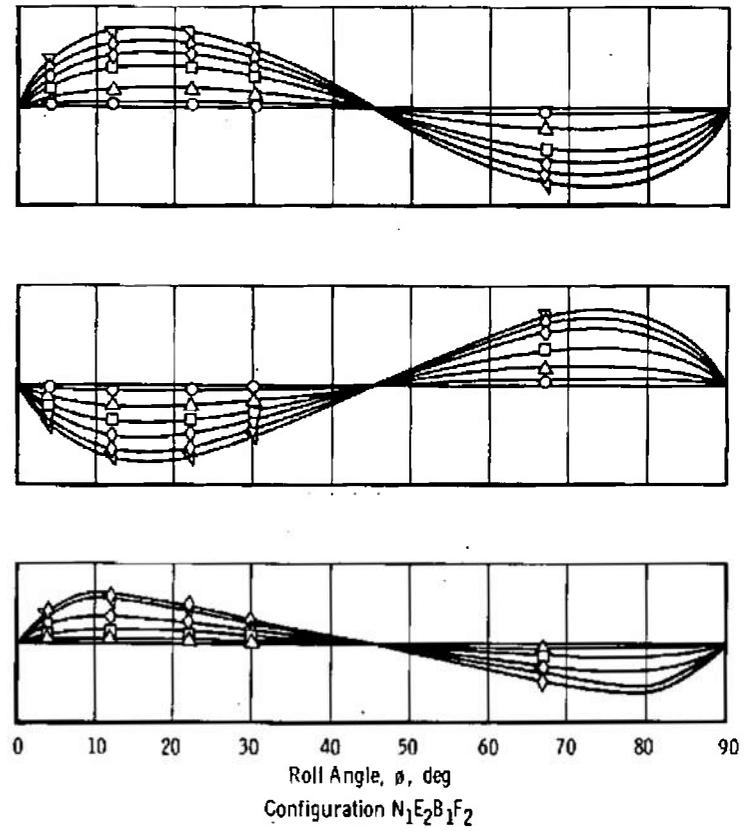
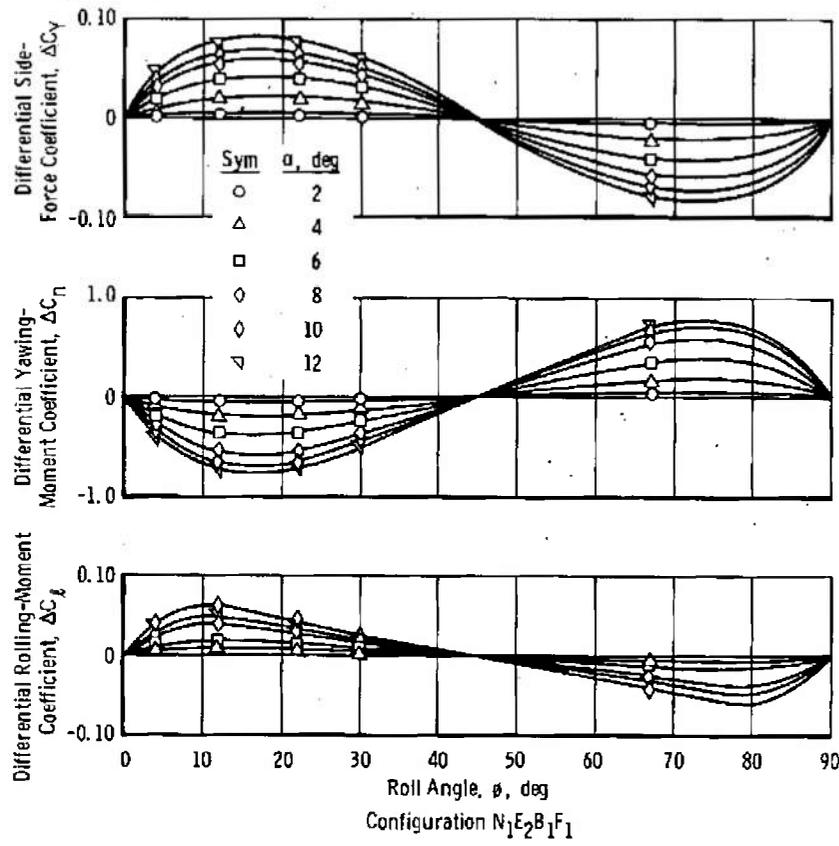


Fig. 7 Effect of Angle of Attack on Lateral Stability Characteristics of Configurations  $N_1E_2B_1F_1$  and  $N_1E_2B_1F_2$  at Mach-6,  
 $Re_\infty/ft = 2.5 \times 10^6$

TABLE I  
TEST SUMMARY

Configuration	$M_\infty$	$Re_\infty \times 10^{-6}, \text{ft}^{-1}$	$\alpha, \text{deg}$	$\phi, \text{deg}$
$N_1E_2B_1F_1$	6.06	5.0	-2 to 14	0 and 12
"	6.03	2.5		0, 4, 12, 22, 30, 45, 67, 90
$N_1E_1B_1F_1$	↓ 7.99 ↓	↓	↓	0
$N_1E_3B_1F_1$				
$N_2E_2B_1F_1$				
$N_3E_2B_1F_1$				0, 4, 12, 22, 30, 45, 67, 90
$N_1E_2B_1F_2$				0
$N_1E_2B_1F_3$				
$N_1E_2B_1$				
$N_1E_1B_1F_1$				
$N_1E_2B_1F_1$				
$N_1E_3B_1F_1$				
$N_2E_2B_1F_1$				
$N_3E_2B_1F_1$				
$N_1E_2B_1$				

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14. KEY WORDS	LINK A		LINK B		LINK C	
	ROLE	WT	ROLE	WT	ROLE	WT
Tomahawk static stability hypersonic flow						

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