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Wolf

Research and Development Corporation

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OPTICAL GENERATOR PROGRAM

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I INTRODUCTION

The program described herein was devised primarily to provide acquisition data to a ground based instrumentation net consisting mainly of various types of cameras. The cameras, in this case, are employed to photograph a flashing strobe light that is mounted on each pole of the ANNA geodetic satellite that is magnetically stabilized along its spin axis.

Provisions have been made to compute a satellite ephemeris by a differential correction procedure as the satellite is stepped around the orbit in some desired increment of time.

It is assumed that the density distribution of the Earth is axially symmetric and that the force field is represented by the principle term and the zonal terms 2 through 4. Provision is also made for accommodating the parameters of a model atmosphere. At each step computations are made to determine:

1) If a given observing site is in darkness (elevation angle to the Sun less than some desired ϵ).
2) If the elevation angle from the observing site to the satellite is positive or greater than some desired ϵ'.
3) The components of the magnetic field (North, East, vertical, horizontal and total field) and the dip and declination.
4) Which strobe, if either, is visible to an observing site.
5) If the recorded image size on the photographic plate will be larger than some ϵ''.

If the above conditions are satisfied, the program continues and computes the following additional information for each observing site:

1) Time of the observation.
2) The azimuth and range and the topocentric hour angle and declination to the satellite.
3) The latitude and longitude of the sub-satellite point.

4) The angle between the observer-satellite vector and the observer-moon vector.

5) If the satellite is illuminated by the Sun.

6) The angle between the observer-satellite vector and the center of the light cone.

Other features of the program are:

1) The selection of the most valuable observations for a station to make from series of possible observations. This is achieved by considering the azimuths at which the station has previously recorded data. On this basis, a final selection of flash times and the associated "look angles" from each site is made.

2) The preparation of teletype messages (see Appendix 1).

3) The capability to compute acquisition data for observers not concerned with the light (range or range-rate stations).

4) A station designated as a "share station" will not have flashes scheduled specifically for it. However, acquisition data will be computed for any scheduled flashes that may be observed by the "share station."

II EPHEMERIS

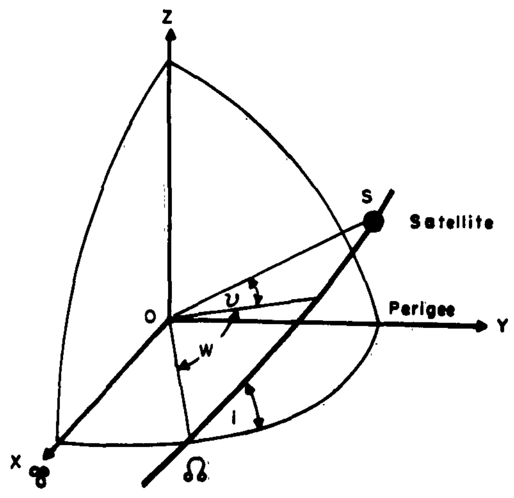


Figure 1

The standard elements (Figure 1)

a = semi major axis

e = eccentricity

i = inclination

Ω = right ascension of the ascending node

ω = argument of perigee

M = mean anomaly

ν = true anomaly

u = $\omega + \nu$, the argument of latitude

are used in the computation of the ephemeris. Time is considered the independent variable and, as the satellite is stepped around the orbit, computations are made to determine the perturbative effects of the Earth's oblateness and drag on the satellite's position.

The radial, transverse, and normal perturbative accelerations (see Figure 2) due to oblateness are:

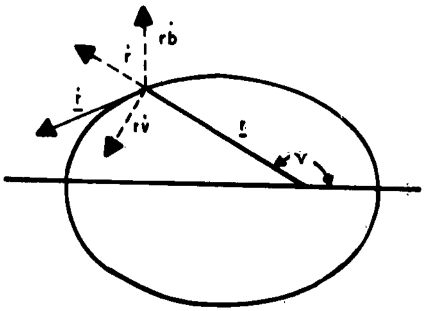


Figure 2

$$\dot{r}'_B = -\frac{\mu}{r^2}\left\{\frac{J}{r^2}\left[1 - \frac{3}{2}\sin^2 i\,(1 - \cos 2u)\right]\right.$$

$$+ \frac{4H}{5r^3}\sin i\left[3(1 - \frac{5}{4}\sin^2 i)\sin u + \frac{5}{4}\sin^2 i \sin 3u\right]$$

$$+ \frac{K}{r^4}\left[\frac{1}{2}(1 - 5\sin^2 i + \frac{35}{8}\sin^4 i) + \frac{5}{2}\cos 2u \sin^2 i\,(1 - \frac{7}{6}\sin^2 i)\right.$$

$$\left.\left. + \frac{35}{48}\cos 4u \sin^4 i\right]\right\}$$

$$r\dot{v}'_B = \frac{\mu}{r^2}\left\{-\frac{J}{r^2}\sin^2 i \sin 2u + \frac{3H}{5r^3}\sin i\left[(1 - \frac{5}{4}\sin^2 i)\cos u + \frac{5}{4}\sin^2 i \cos 3u\right]\right.$$

$$\left. - \frac{K}{r^4}\sin^2 i\left[\sin 2u\,(1 - \frac{7}{6}\sin^2 i) + \frac{7}{12}\sin^2 i \sin 4u\right]\right\}$$

$$r\dot{b}'_B = \frac{\mu}{r^2}\left\{-\frac{2J}{r^2}\sin i \cos i \sin u + \frac{3H}{5r^3}\cos i\left[1 - \frac{5}{2}\sin^2 i\,(1 - \cos 2u)\right]\right.$$

$$\left. - \frac{K}{r^4}\sin i \cos i\left[2\sin u\,(1 - \frac{7}{4}\sin^2 i) + \frac{7}{6}\sin^2 i \sin 3u\right]\right\}$$

The radial, transverse and normal perturbative accelerations due to drag are:

$$\dot{r}_D^{\backprime} = \frac{A}{m} \rho a e V \sin E \dot{E}$$

where:

$$\dot{E} = \frac{n}{1 - e \cos E}$$

$$\rho = \text{air density}$$

$$r\dot{v}_D^{\backprime} = - \frac{A}{m} \rho a (1 - e^2)^{\frac{1}{2}} V \left[1 - d (1 - e \cos E)^2/(1 - e^2) \right] \dot{E}$$

$$r\dot{b}_D^{\backprime} = - \frac{A}{m} \rho a \omega_s \left(\frac{1 - e \cos E}{n} \right)^2$$

where:

$$V = \left(\frac{\mu}{a} \right)^{\frac{1}{2}} \left(\frac{1 + e \cos E}{1 - e \cos E} \right)^{\frac{1}{2}} \left[1 - d \left(\frac{1 - e \cos E}{1 + e \cos E} \right) \right]$$

$$d = \left(\frac{\omega_s}{n} \right) (1 - e^2)^{\frac{1}{2}} \cos i$$

$$\omega_s = \text{rotational rate of the Earth}$$

Expressions for the perturbative effect on the elements.

$$a^{\backprime} = - \frac{2}{3} a \frac{n^{\backprime}}{n}$$

$$\frac{n^{\backprime}}{n} = - \frac{3}{1 - e^2} \left[\frac{\dot{r}^{\backprime}}{\sqrt{\mu p}} e p \sin \nu + \frac{r\dot{v}^{\backprime}}{\sqrt{\mu p}} \frac{p^2}{r} \right]$$

where

$$p = a(1 - e^2)$$

$$e^{\backprime} = \frac{r \dot{r}^{\backprime}}{\sqrt{\mu p}} \left(\frac{p}{r} \sin \nu \right) + \frac{r^2 \dot{v}^{\backprime}}{\sqrt{\mu p}} \left[\left(\frac{p}{r} + 1 \right) \cos \nu + e \right]$$

$$i^{\backprime} = \frac{r^2 \dot{b}^{\backprime}}{\sqrt{\mu p}} \cos u$$

$$\Omega^{\backprime} = \frac{r^2 \dot{b}^{\backprime}}{\sqrt{\mu p}} \frac{\sin u}{\sin i}$$

$$\omega^{\backprime} = u^{\backprime} - \nu^{\backprime}$$

$$u^{\backprime} = -\Omega^{\backprime} \cos i$$

$$\nu^{\backprime} = \frac{1}{e}\left[\frac{r\dot{r}^{\backprime}}{\sqrt{\mu p}}\left(\frac{p}{r}\cos\nu\right) - \frac{r^2\dot{v}^{\backprime}}{\sqrt{\mu p}}\left(\frac{p}{r}+1\right)\sin\nu\right]$$

$$E^{\backprime} = \left(1 - e^2\right)^{\frac{1}{2}}(u^{\backprime} - \omega^{\backprime}) - \frac{a^{\backprime} e}{2a} \sin E - \frac{\dot{r}^{\backprime} r \dot{v}^{\backprime}}{\sqrt{\mu a}}$$

Perigee height, $h_\pi = a(1 - e) - Re_\pi$

Radial distance, $r = a(1 - e \cos E)$

The difference between the height of the satellite above the Earth (h'') at any time and the height of the satellite above the Earth at perigee $\left(h_\pi\right)$

$$h'' - h_\pi = (Re_\pi - Re_{h''}) + a e\left[1 - \cos E\right]$$

The air density, $\rho = \rho_\pi e^{-Kdh}$

where

$$K = \frac{1}{h} = \frac{1}{\text{scale height}}$$

ρ_π = air density at perigee

The total perturbative effect of bulge and drag on the elements at any time

$$a = a' + (a_D + a_B)\, dt$$

$$e = e' + (e_D + e_B)\, dt$$

$$i = i' + (i_D + i_B)\, dt$$

$$\Omega = \Omega' + (\Omega_D + \Omega_B)\, dt$$

$$\omega = \omega' + (\omega_D + \omega_B)\, dt$$

$$E = E' + (\dot{E} + E_D + E_B)dt$$

$$u = u' + (\dot{u} + u_D + u_B)dt$$

where:

$$\dot{E} = \frac{n}{1 - e \cos E}$$

$$\dot{u} = \frac{\sqrt{1 - e^2}}{1 - e \cos E} \dot{E}$$

Dots denote the two body changes.

Primes denote the value at the previous step.

III ACQUISITION DATA

At each step values of a, e, i, Ω ; u and ν are computed after which the program enters the "acquisition data" phase. The computations for this phase break into two groups: those which are made only once and those which are made once for each observation station.

When a coordinate system is not specified in what follows, an "Earth-fixed" system is to be assumed, i.e., $\underline{i}$ and $\underline{j}$ in the equatorial plane with $\underline{i}$ pointed toward Greenwich, $\underline{k} = \underline{i} \times \underline{j}$.

Group 1: Computations made only once

1) Compute λ_n = west longitude of node.

a) θ_G = right ascension of Greenwich.

$$= \theta_{Go} + (t - t_o)\dot{\theta}_{1\,days} + (t - t_o)\dot{\theta}_{2\,fract}$$

b) $\lambda_n = \theta_G - \Omega$ $\qquad -\pi < \lambda_n \leq \pi$

2) Compute λ_s, west longitude of sub-satellite point.

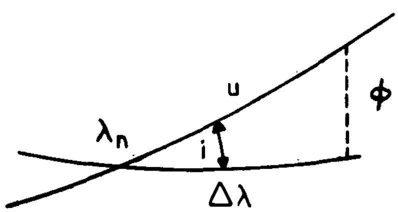


Figure 3

$$\Delta\lambda = \tan^{-1}\left(\frac{\cos i}{\cot u}\right)$$

$$\lambda_s = \lambda_n - \Delta\lambda \qquad -\pi < \lambda_s \leq \pi$$

3) Compute ϕ_s, the geodetic latitude of sub-satellite point.

δ = declination of satellite

$$\tan\delta = \frac{\sin i \sin u}{\sqrt{1 - \sin^2 i \sin^2 u}}$$

$$\tan\phi_s = \frac{1}{1 - \tilde{e}^2} \tan\delta; \quad \tilde{e}^2, \text{ spheroid eccentricity} = .0067686579$$

$$\phi_s = \tan^{-1}\left\{\left(\frac{1}{1 - \tilde{e}^2}\right) \tan\delta\right\}$$

4) Compute r, distance from center of Earth to satellite.

$$r = a(1 - e\cos E)$$

Group 2: Computations made for each station at each time step

5) Elevation angle to the Sun

$$\text{Elev} = \sin^{-1}\left[\sin\phi \sin\delta_\odot + \cos\phi \cos\delta_\odot \cos\beta\right]$$

where ϕ = observers latitude

$\delta_\odot$ = declination of the Sun

β = LST - $\alpha_\odot$

LST = local sidereal time

$\alpha_\odot$ = right ascension of the Sun

where

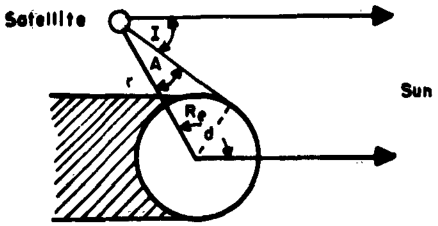
a) $\alpha_\odot = \ell_\odot - C_{16} \sin 2\ell_\odot$; $\delta_\odot = C_{17} \sin\alpha_\odot$

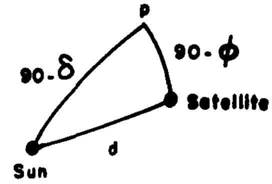
b) $L_\odot$ = longitude of Sun at t_o (Jan 0. 0 of year of interest)

c) $\ell_\odot = L_\odot + C_2 (t - t_o) + C_{15} \sin (C_2 [t - t_o] - C_{14})$

d) $\phi_\odot = \tan^{-1}\left\{\frac{1}{1 - \tilde{e}^2} \delta_\odot\right\}$; $\lambda_\odot = \theta_G - \alpha_\odot$

6) **If the satellite is illuminated by the Sun, the angle I must be positive (Figure 4a).**





I = 180 - d - A

$A = \sin^{-1}\left(\frac{Re}{r}\right)$

$d = \cos^{-1}(\sin\phi_{\odot}\sin\phi_{\circ} + \cos\phi_{\odot}\cos\phi_{\circ}\cos\beta)$

where: $_{\circ}$ denotes satellite

$\beta = |RA_{\odot} - RA_{\circ}|$

7) Magnetic dip and declination are computed by evaluating the gradient of the magnetic potential at altitude (details in Appendix II) to obtain the x, y and z components of the magnetic field. These quantities permit the computation of (Figure 5).

a) Horizontal component of Earth's magnetic field (H)
b) Total field vector (F)
c) Vertical component of Earth's magnetic field (V)
d) Horizontal component of Earth's magnetic field (H)
e) Total field vector (F)

Magnetic dip, $a_m = \cos^{-1}\left[\frac{H}{F}\right]$

Magnetic declination, $\delta_m = \cos^{-1}\left[\frac{Z}{H}\right]$, plus Easterly

8) The light angle, Θ, (Figure 6) is the angle between the station-satellite vector and a unit vector in the direction of the center of the light cone. Orientation of $\hat{S}$, a unit vector in direction of the North axis of the light cone as a function if its latitude (ϕ), longitude (λ), magnitude dip (a_m) and magnetic declination (δ_m), is

$$\hat{S} = \underline{i}\left\{\cos a_m \sin\delta_m \sin\lambda - \cos\lambda\left[\cos\delta_m \cos a_m \sin\phi + \sin a_m \cos\phi\right]\right\}$$

$$+ \underline{j}\left\{\cos a_m \sin\delta_m \cos\lambda + \sin\lambda\left[\cos\delta_m \cos a_m \sin\phi + \sin a_m \cos\phi\right]\right\}$$

$$+ \underline{k}\left\{\cos\delta_m \cos a_m \div \sin a_m \sin\phi\right\}$$

$$\theta = \cos^{-1}\left\{\frac{\underline{\underline{ST}}\cdot\hat{S}}{|ST|}\right\}$$

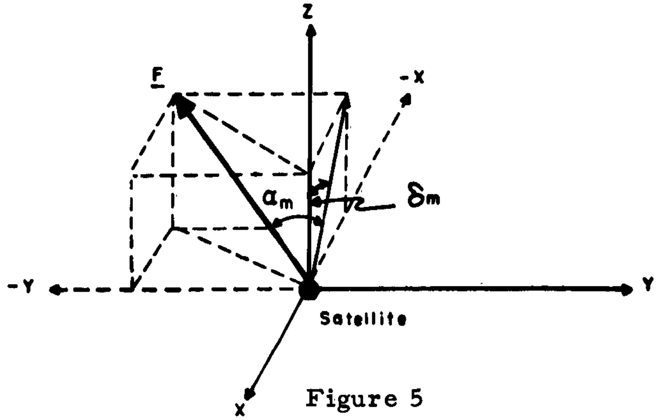


Figure 5

A vector from the center of the Earth to the satellite, $\underline{OS}$, is

$$\underline{OS} = r\left[(\cos\phi_s \cos\lambda_E)\,\underline{i} + (\cos\phi_s \sin\lambda_E)\,\underline{j} + \sin\phi_s\ \underline{k}\right]$$

$$\underline{OT} = (R_e + h)\left[(\cos\phi_T \cos\lambda_{E_T})\,\underline{i} + (\cos\phi_T \sin\lambda_{E_T})\,\underline{j} + \sin\phi_T\,\underline{k}\right]$$

$$\underline{OS} - \underline{OT} = \underline{TS}$$

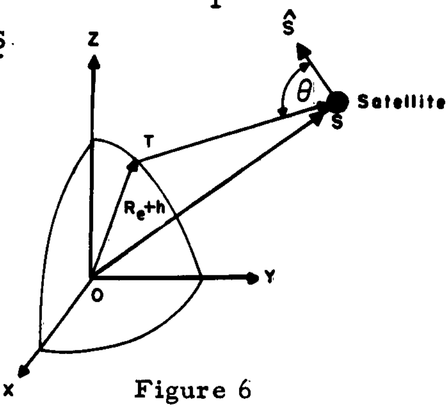


Figure 6

8) Elevation angle to the satellite

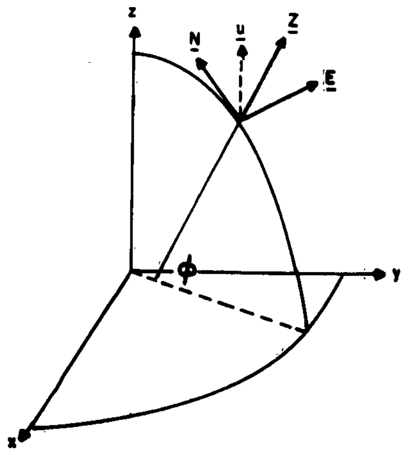


Figure 7

Define a unit vector $\underline{Z}$ normal to the spheroid at the observer (Figure 7)

$$\underline{Z} = \cos\phi \; \cos\lambda_E \underline{i} + \cos\phi \; \sin\lambda_E \underline{j} + \sin\phi \; \underline{k}$$

A vector $\underline{N}$, in the meridian and pointing North

$$\underline{N} = -\sin\phi \; \cos\lambda_E \underline{i} + \sin\phi \; \sin\lambda_E \underline{j} + \cos\phi \; \underline{k}$$

$$\underline{E} = \underline{N} \times \underline{Z}$$

$\underline{u}$ is a unit vector from the station to the satellite

$$\underline{u} = \frac{\underline{\underline{TS}}}{|\underline{\underline{TS}}|} \qquad \text{(see Figure 6)}$$

$$\text{Elev Angle} = \sin^{-1} (\underline{Z} \cdot \underline{u})$$

10) Azimuth from the station to the satellite

$$\text{Azimuth} = \cos^{-1} \left\{ \frac{\underline{Z} \times \underline{u}}{|\underline{Z} \times \underline{u}|} \cdot (-\underline{E}) \right\}$$

11) Topocentric hour angle and declination.

$\underline{L}$, a unit vector from the station to the satellite in inertial coordinates is

$$\underline{L} = x\underline{i} + y\underline{j} + z\underline{k}$$

$$\text{Declination} = \sin^{-1}(z)$$

$$\text{Right ascension} = \tan^{-1}\left(\frac{y}{x}\right)$$

$$\text{Hour angle} = \text{LST} - \text{RA}$$

Some camera stations require the hour angle and declination of each flash. Since the ephemeris is stepped in time by an amount not equal to the time interval between flashes, but by a multiple of the interval between flash sequences, the time derivatives of x, y, and z are computed and multiplied by the time between flash intervals to obtain the new coordinates of each flash. With these new coordinates, the right ascension and declination are then computed as noted above.

The time derivatives are

$$\dot{x} = \dot{r}[1 - \sin^2 i \sin^2 u]^{\frac{1}{2}} \cos\lambda +$$

$$r[(- \sin i \cos i \sin^2 u \dot{i} + \sin^2 i \sin u \cos u \dot{u})/(1 - \sin^2 i \sin^2 u)^{\frac{1}{2}} \cos\lambda] +$$

$$r[(1 - \sin^2 i \sin^2 u)^{\frac{1}{2}} \sin\lambda/(\cos^2 u \cos^2 i \sin^2 u)(\cos i \dot{u} -$$

$$\sin i \sin u \cos u \dot{i}) + \dot{\Omega} - \omega_s]$$

$$\dot{y} = -\dot{r}[1 - \sin^2 i \sin^2 u]^{\frac{1}{2}} \sin\lambda -$$

$$r[- \sin i \cos i \sin^2 u \dot{i} + \sin^2 i \sin u \cos u \dot{u})/(1 - \sin^2 i \sin^2 u)^{\frac{1}{2}} \sin\lambda] +$$

$$r[(1 - \sin^2 i \sin^2 u)^{\frac{1}{2}} \cos\lambda)/(\cos^2 u \cos^2 i \sin^2 u)(\cos i \dot{u} - \sin i \sin u \cos u \dot{i}) +$$

$$\dot{\Omega} - \omega_s]$$

$$\dot{z} = \dot{r} \sin i \sin u + r \cos i \sin u \dot{i} + r \sin i \cos u \dot{u}$$

12) Image size of the flash

Image diameter (in microns) = $a_1 q + a_2 q^2 + a_3 q^3$

where a_1, a_2, a_3 are functions of the plate emulsion. For 103F emulsion developed 8 minutes $a_1 = 7.468$, $a_2 = -0.112237$, $a_3 = .0008352$

$$q = \left(\frac{D}{s}\right) (T B)^{\frac{1}{2}} e^{-0.46 \Delta m / \sin h}$$

D = aperture of camera

s = distance from satellite to observer = $|\underline{TS}|$

T = transmission of lens

B = 9000 if $\theta \leq 45°$

$150(105 - \theta_i)$ if $\theta > 45°$

Δm = atmospheric extinction at the zenith for the station; when an actual measurable value is unavailable, $\Delta m = 1.25$ (the factor for moderate haze).

h = elevation angle

13) Right ascension, α_m, and declination, δ_m, of the moon are obtained from a lunar ephemeris listed in memory. α_m and δ_m are given for 0 hours U. T. of each day of the year. A six point interpolation yields the desired quantities at any time, t. For purposes of this program the vector from the center of the Earth to the Moon is taken as being the same as the vector from the observer to the Moon. The angle between the station-satellite vector and the station-moon vector, $\underline{TM}$, is:

Moon Angle = $\cos^{-1}\left(\underline{TS} \cdot \underline{TM}\right)$

Moon Phase, P_m, is determined from

$$P_m = R.A._m - R.A._\odot$$

if $P_m = 0° \pm 45°$, Phase new

$P_m = 180° \pm 45°$, Phase full

Otherwise, Phase quarter

14) Observation Selection. From a series of possible observations, one observation (the least valuable one) is discarded. If a station has made n good observations at azimuths A_1, A_2, A_3 -- A_n and it is now possible to make an observation at azimuth B, a "value" is assigned to azimuth B in the following fashion:

form $$\bar{A} = \frac{1}{n} \sum_{i=1}^{n} A_i$$

and

$$p = \frac{1}{n} \sum_{i=1}^{n} (A_i - \bar{A})^2$$

take as the "value" of the observation

$$V(B) = e^{-p \sin^2\left(\frac{\bar{A} - B}{2}\right)}$$

Note that $0 \leq V(B) \leq 1$, that $V(B)$ is zero if $\bar{A} \approx B$, and that it also is small if p is large (that is, there is a large "scatter" in the azimuths already observed). The values n, p, and $\bar{A}$ are "updated" as additional good observations are made.

INPUT AND OUTPUT FORMATS

A. <u>For Determining Acquisition Data and Selecting Observations</u>

The OGP program uses logical tapes 8 and 11 for input. Logical tapes 9, 10, 12 and 14 are used as intermediate binary tapes.

The acquisition data and selected observations are available as printed output on logical tape 5 with data select zero. With data select 4 the teletype messages to APL, NASA and the 1381st Geodetic Survey Squadron are punched from tape 5. (Teletype messages are also available as printed output on logical tape 6 with data select zero). Control cards necessary for this computation are input cards 1 through 6 (Table I) and four station cards for each observing site involved (Table II). Note that in the preparation of station cards (Table II) a station may be designated as a "Share Station."

B. <u>For Ephemeris Computation Only</u>

If so desired, the ephemeris portion only is available in the form of latitude, longitude and time. In this instance the first six control cards must be followed by four blank cards.

TABLE I

INPUT FORMAT – ELEMENT CARDS

Card No.	Card Column	Format	Contents
1	1-6	12A6	Year, month, day of this run
	7-72		Job heading
2	1-12	F12.9	EPOCHT (days and decimals of day)
	13-16	I4	IYEAR (year of epoch)
	17-22	F6.3	CONANG (1/2 light angle in deg.)
	23-28	F6.3	SUNTST (Test for sun elev. in deg.)
	35-40	F6.3	DELTAE (ΔE in deg.)
	42-56	F15.15	QJ2 (J_2)
	57-71	F15.15	QJ3 (J_3)
3	1-10	F10.6	WO (ω_o in deg.)
	12-21	F10.6	RAO (Ω_o in deg.)
	23-32	F10.5	AXIS (semi-major axis in nautical miles)
	34-43	F10.9	ECCNO (eccentricity)
	45-54	F10.8	XINC (inclination in degrees)
	56-61	F6.3	QMASS (mass of satellite in kilograms)
	63-72	F10.4	AREA (cross sectional area of satellite in square centimeters)
4	1-5	I5	NSTA (the number of stations to be considered)
	6-10	I5	NREV (number of revolutions to be considered for this run)
	15-25	F11.7	DM1 (M_o the initial mean anomaly in degrees)
5	1-10	I10	IRM (the maximum number of flash sequences to be allowed per revolution)
	11-20	I10	ILM (the maximum number of flash sequences to be allowed per load)
	23-34	F12.9	GDT (integer by which the interval between clock pulses is multiplied to determine time interval at which acquisition date is computed; the elements are integrated at one half this interval)

40-49	I10	INF (initial number of flash sequences executed to date)
50-59	I10	INFL (initial number of injections executed to date)
60-71	F12.9	TIMP (epoch of an even flash time)
72	I1	Blank if refined projections - "one" if long range (an identifier for message to NASA)
1-12	F12.9	DDTT (time in seconds between flashes in a sequence)
13-24	F12.9	TSTOP (stop time in days)
25-28	F4.1	Minimum angle between station-satellite vector and station new moon vector in degrees
29-32	F4.1	Minimum angle between station-satellite vector and station quarter moon vector in degrees
33-36	F4.1	Minimum angle between station-satellite vector and station full moon vector in degrees
39-50	F12.9	ELTIM (time in days at which acquisition data computations begin)
53-57	F5.2	Minimum image size in microns
60	I1	Blank for acquisition data; 1 for ephemeris computation only

TABLE II

STATION CARDS

Card No.	Card Column	Format	Contents
1	1-24	4A6	Station number and name (number in col. 1-4 with leading zeros)
	25-34	F10. 5	XLAT (station latitude in degrees)
	35-44	F10. 5	XLONG (Station west longitude in degrees)
	45-54	F10. 2	HEIGHT (height of station above MSL in feet)
	70	I1	"1" if this is MOTS station
	72	I1	"1" if this is camera station
			"2" if this is range station
			"3" if this is range rate station
2	1	I1	"1" if this station can only share flashes
	11-20	F10. 7	DIAM (camera aperture in mm)
	21-30	F10. 7	TRANS (lens transmission factor)
	31-40	F10. 7	QM (atmospheric extinction at observer zenith
	41-50	F10. 7	A1 } Constants relative to film type and
	51-60	F10. 7	A2 } development procedure
	61-70	F10. 7	A3
3	1-15	F15. 10	C (Average azimuth of all flashes observed from station)
	16-30	F15. 10	D (Weight factor for selecting observations)
4			Lowest elevation angle (degrees) attainable as a function of azimuth
	1-3	(I3)	Elevation Angle 0°-20° Azm (0°N)
	4-6	(I3)	" " 20-40
	7-9	(I3)	" " 40-60
	10-12	(I3)	" " 60-80
	13-15	(I3)	" " 80-100
	16-18	(I3)	" " 100-120
	19-21	(I3)	" " 120-140

22-24	(I3)	**Elevation**	**Angle**	140-160
25-27	(I3)	"	"	160-180
28-30	(I3)	"	"	180-200
31-33	(I3)	"	"	200-220
34-36	(I3)	"	"	220-240
37-39	(I3)	"	"	240-260
40-42	(I3)	"	"	260-280
43-45	(I3)	"	"	280-300
46-48	(I3)	"	"	300-320
49-51	(I3)	"	"	320-340
52-54	(I3)	"	"	340-0

GLOSSARY OF SYMBOLS USED

A	Cross sectional area of satellite in square centimeters
C_{14}	(longitude of perigee of sun-longitude of sun at January 0. 0 for year of interest)
C_{15}	2 (eccentricity of earth's orbit)
C_{16}	$\tan^2$ (mean obliquity of ecliptic/2)
C_{17}	tan (mean obliquity of ecliptic)
J	1623×10^{-6}
H	-6×10^{-6}
K	9×10^{-6}
$\ell_{\odot}$	longitude of sun at any time
m	mass of satellite in kilograms
n	Mean motion of satellite $(GM)^{1/2}/a^{3/2}$
Re_{π}	radius of earth at perigee
$Re_{h''}$	radius of earth at sub-satellite position
V	Satellite velocity
$\dot{\theta}_1$	$0^{\circ}.98564724$
$\dot{\theta}_2$	$360^{\circ}.98564724$
μ	GM
ϕ_s	Satellite latitude
λ_E	East longitude

Program Constants and Conversion Factors

↓

READ INPUT CARDS AND STORE OUTPUT TITLES

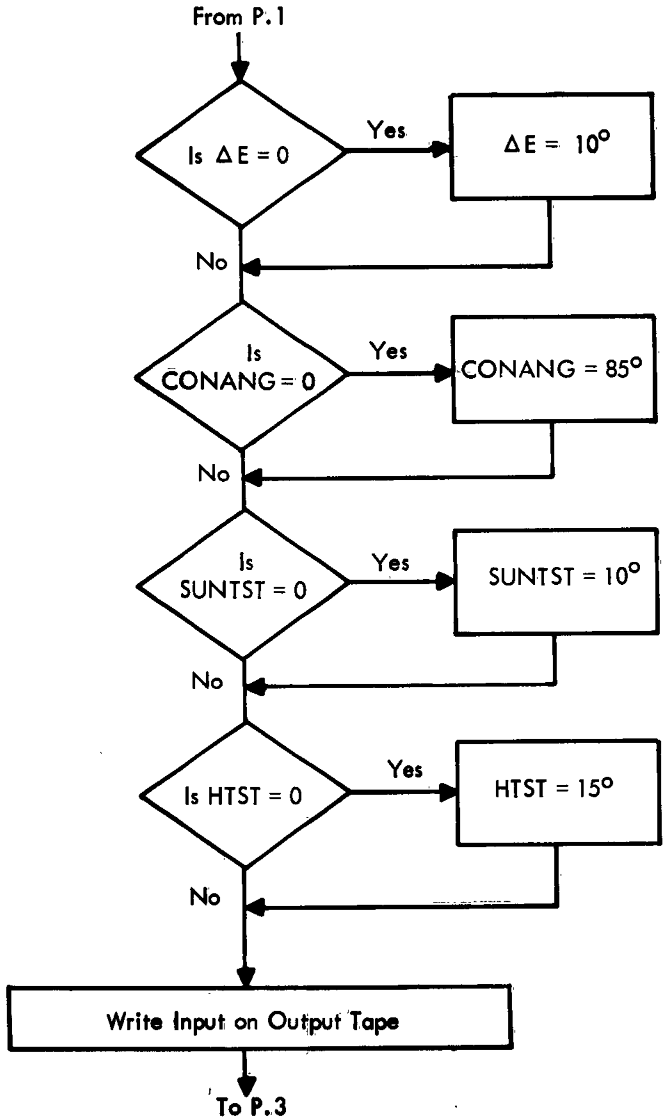
EPOCHT	=	Epoch of Elements in Days
IYEAR	=	Year of Epoch
CONANG	=	1/2 Cone Angle of Strobe Light
SUNTST	=	Maximum Sun Elevation Angle
DELTAE	=	ΔE
QJ2	=	J2
QJ3	=	J3
WO	=	ω_o
RAO	=	Ω_o
AXIS	=	a_o
ECCNO	=	e_o
XINC	=	i_o
QMASS	=	Mass of Satellite
AREA	=	Area of Satellite
NSTA	=	No. of Stations
NREV	=	No. of Revolutions to be Considered
DM1	=	M_o
IRM	=	Max. No. of Flashes/Rev.
ILM	=	Max. No. of Flashes/Load
INFL	=	Initial No. of Flash Loads
TIMP	=	Epoch of an Even Clock Pulse
IP	=	Flag to Define Long Range or Refined Predictions
DDTT	=	Time Between Clock Pulses
TSTOP	=	Final Epoch
QLU1	=	Min. Angle Between Satellite & New Moon
QLU2	=	Min. Angle Between Satellite & Qtr. Moon
Q	=	Min. Angle Betwen Satellite & Full Moon
ELTIM	=	Time Look Angles Start
QIMSZ	=	Min. Image Size

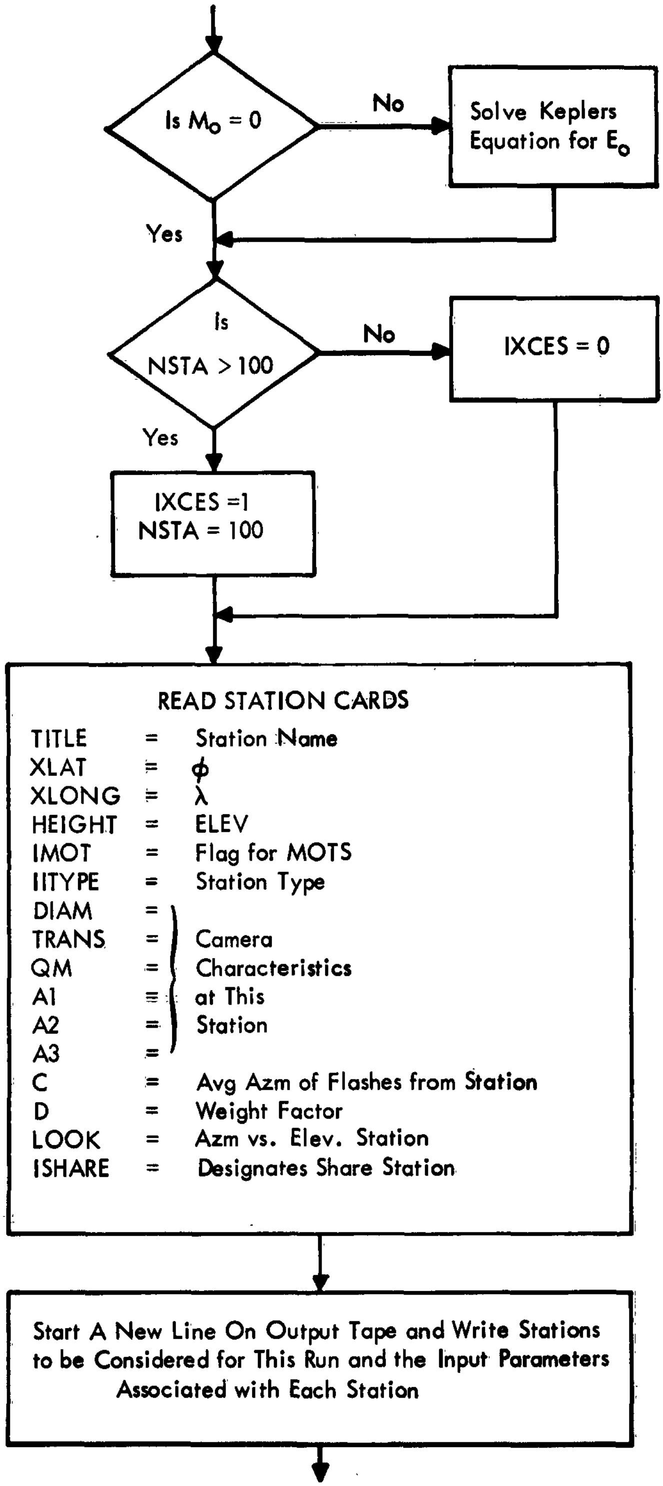
↓

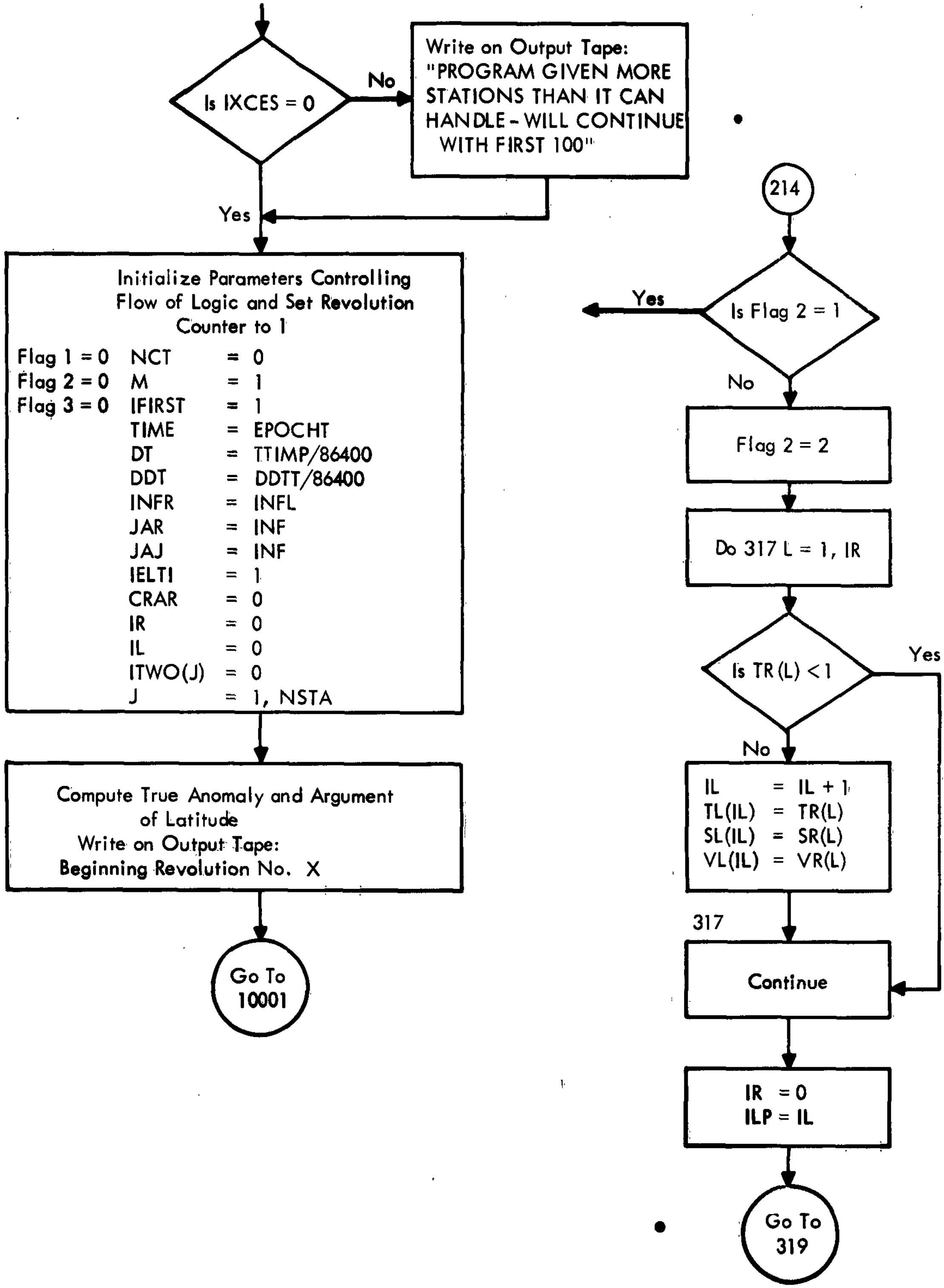
Call CONGET for Annual Constants

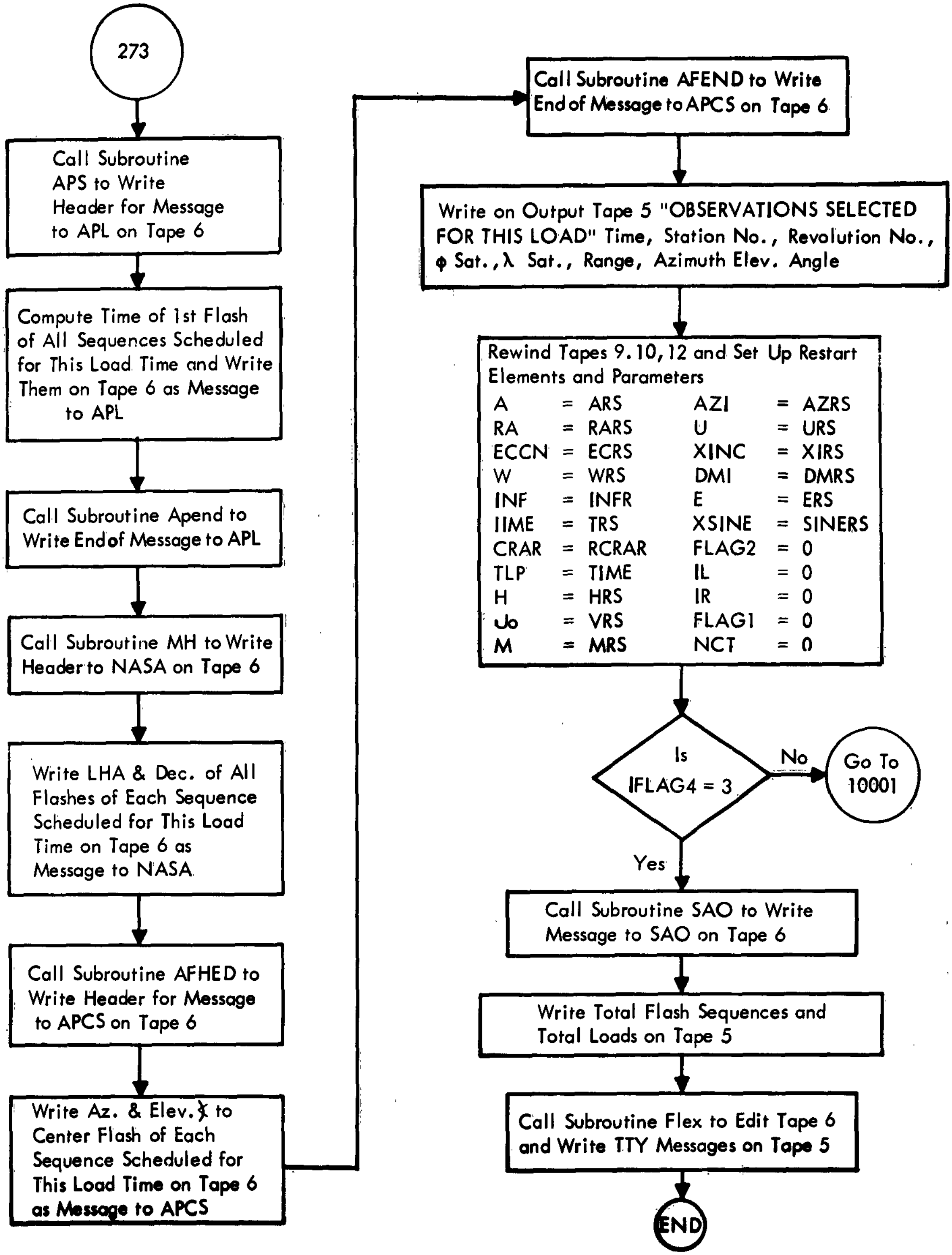
↓

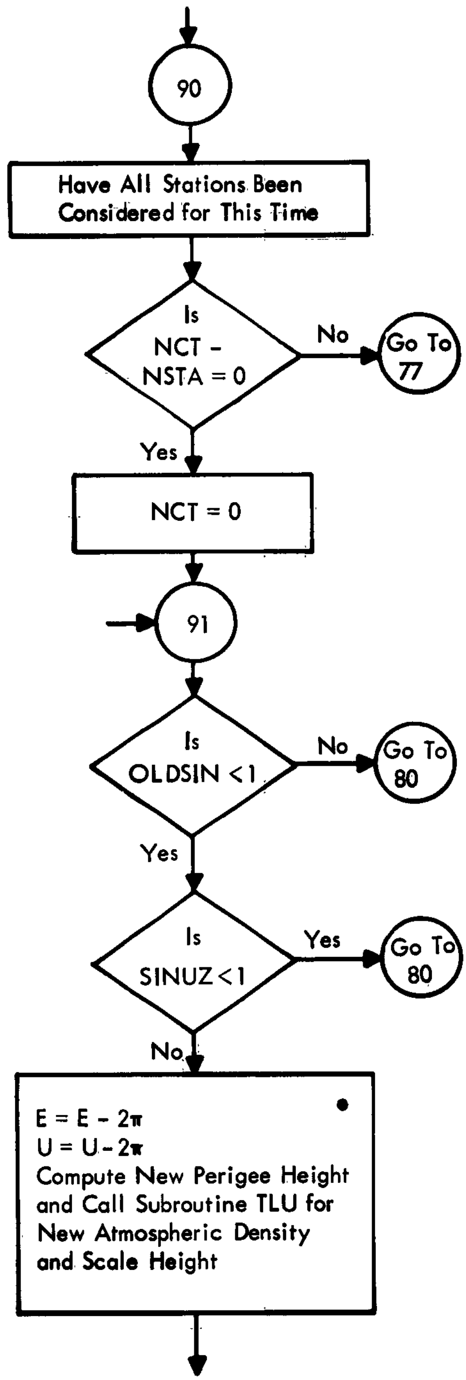
To P.2

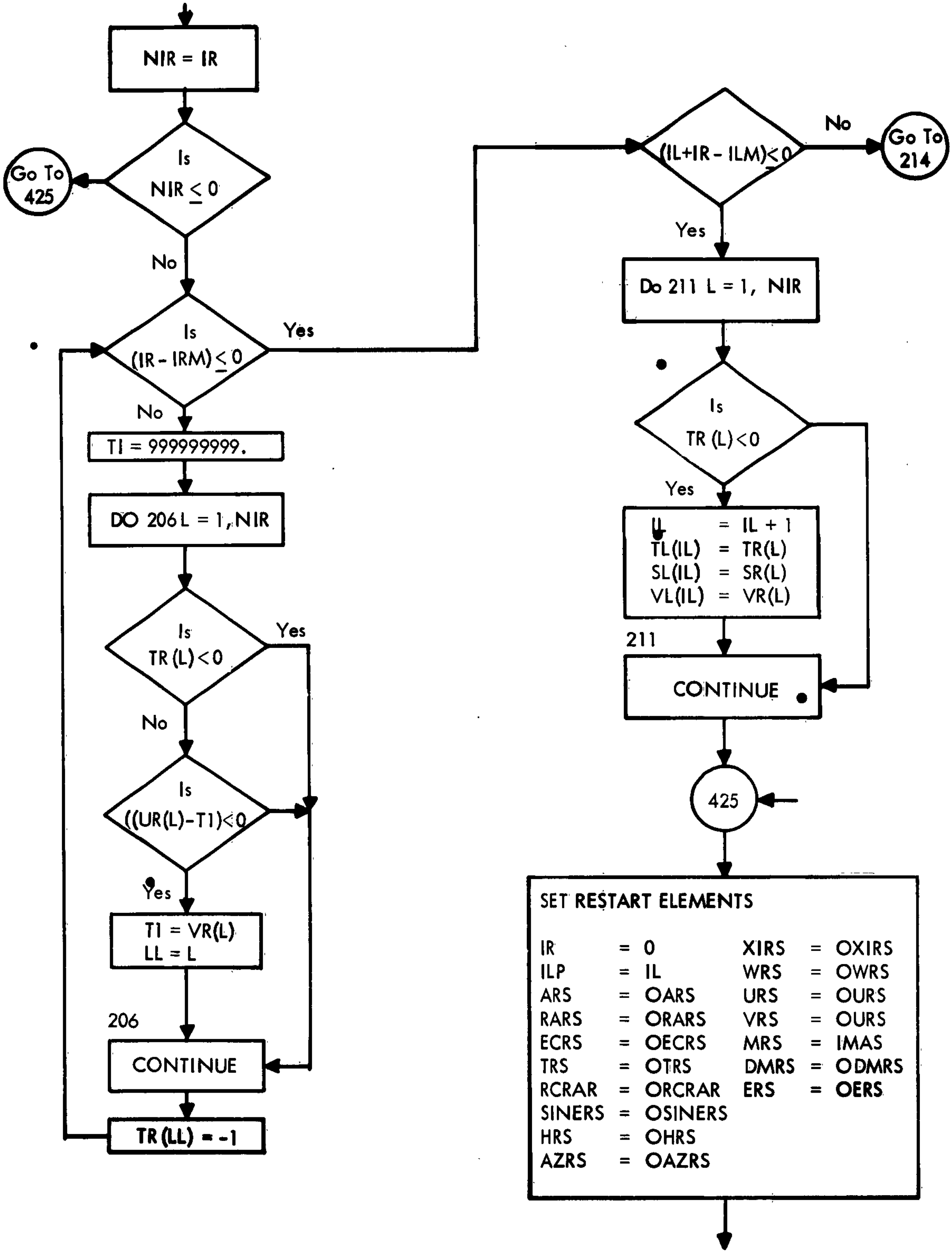

From P. 1
Is $\Delta E = 0$
Yes
$\Delta E = 10^{\circ}$
No
Is CONANG = 0
Yes
CONANG = 85°
No
Is SUNTST = 0
Yes
SUNTST = 10°
No
Is HTST = 0
Yes
HTST = 15°
No
Write Input on Output Tape
To P.3

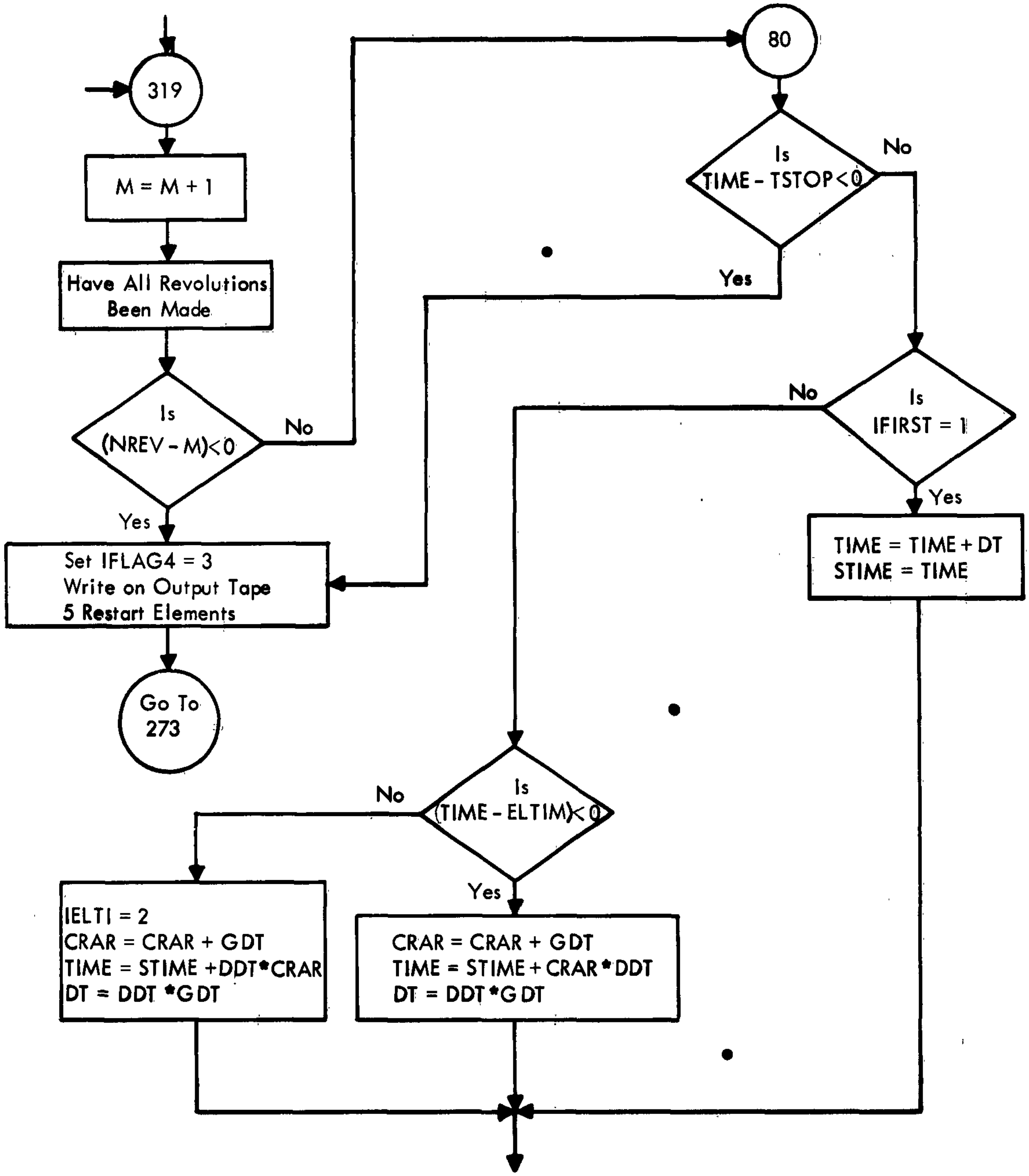

Is $M_o = 0$
No
Solve Keplers Equation for E_o
Yes
Is NSTA > 100
No
IXCES = 0
Yes
IXCES =1
NSTA = 100
READ STATION CARDS
TITLE = Station Name
XLAT = ϕ
XLONG = λ
HEIGHT = ELEV
IMOT = Flag for MOTS
IITYPE = Station Type
DIAM =
TRANS =
QM =
A1 =
A2 =
A3 =
Camera Characteristics at This Station
C = Avg Azm of Flashes from Station
D = Weight Factor
LOOK = Azm vs. Elev. Station
ISHARE = Designates Share Station
Start A New Line On Output Tape and Write Stations to be Considered for This Run and the Input Parameters Associated with Each Station

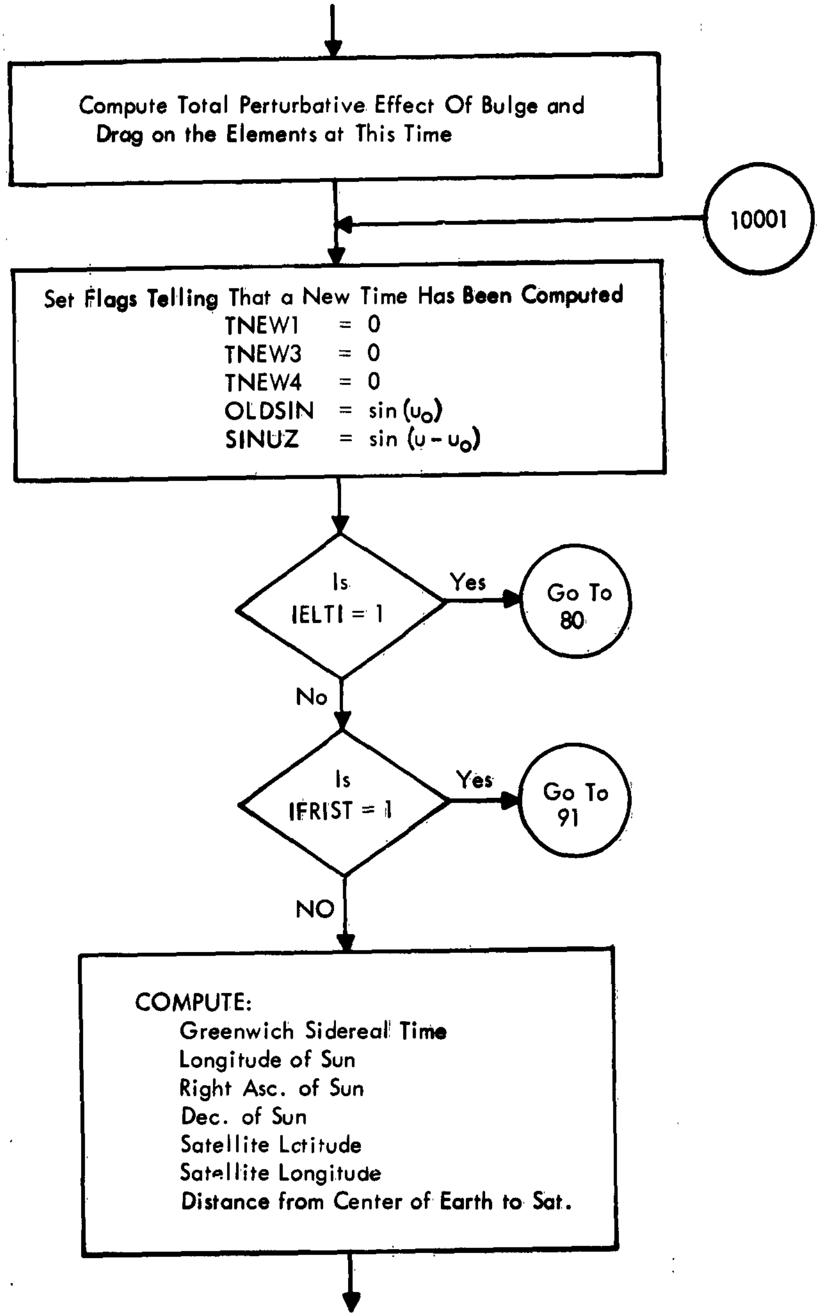

Is IXCES = 0
No
Write on Output Tape: "PROGRAM GIVEN MORE STATIONS THAN IT CAN HANDLE - WILL CONTINUE WITH FIRST 100"
Yes
Initialize Parameters Controlling Flow of Logic and Set Revolution Counter to 1
Flag 1 = 0
Flag 2 = 0
Flag 3 = 0
NCT = 0
M = 1
IFIRST = 1
TIME = EPOCHT
DT = TTIMP/86400
DDT = DDTT/86400
INFR = INFL
JAR = INF
JAJ = INF
IELTI = 1
CRAR = 0
IR = 0
IL = 0
ITWO(J) = 0
J = 1, NSTA
Compute True Anomaly and Argument of Latitude
Write on Output Tape:
Beginning Revolution No. X
Go To 10001
214
Is Flag 2 = 1
Yes
No
Flag 2 = 2
Do 317 L = 1, IR
Is TR(L) < 1
Yes
No
IL = IL + 1
TL(IL) = TR(L)
SL(IL) = SR(L)
VL(IL) = VR(L)
317
Continue
IR = 0
ILP = IL
Go To 319

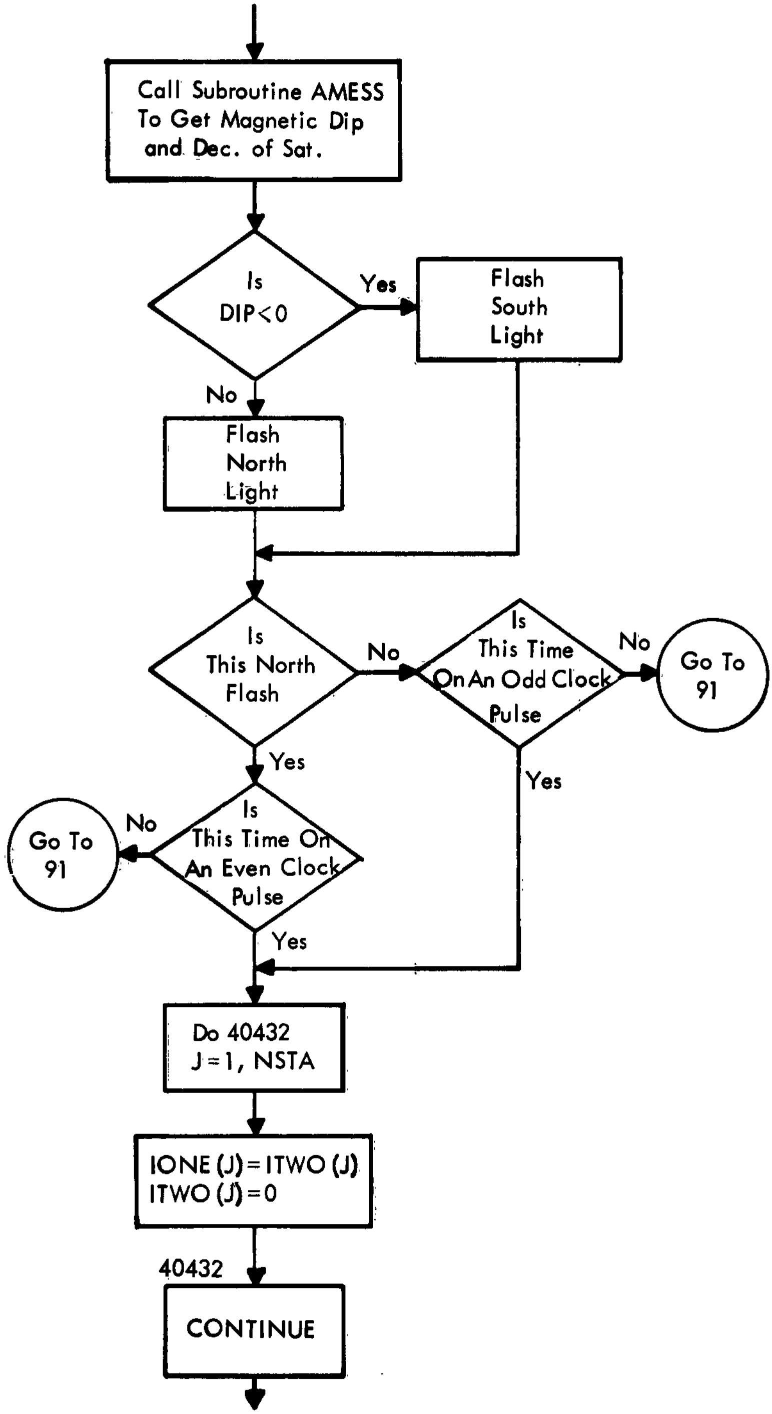

273
Call Subroutine APS to Write Header for Message to APL on Tape 6
Compute Time of 1st Flash of All Sequences Scheduled for This Load Time and Write Them on Tape 6 as Message to APL
Call Subroutine Apend to Write End of Message to APL
Call Subroutine MH to Write Header to NASA on Tape 6
Write LHA & Dec. of All Flashes of Each Sequence Scheduled for This Load Time on Tape 6 as Message to NASA
Call Subroutine AFHED to Write Header for Message to APCS on Tape 6
Write Az. & Elev. ∢ to Center Flash of Each Sequence Scheduled for This Load Time on Tape 6 as Message to APCS
Call Subroutine AFEND to Write End of Message to APCS on Tape 6
Write on Output Tape 5 "OBSERVATIONS SELECTED FOR THIS LOAD" Time, Station No., Revolution No., φ Sat., λ Sat., Range, Azimuth Elev. Angle
Rewind Tapes 9.10, 12 and Set Up Restart Elements and Parameters
A = ARS
RA = RARS
ECCN = ECRS
W = WRS
INF = INFR
IIME = TRS
CRAR = RCRAR
TLP = TIME
H = HRS
Uo = VRS
M = MRS
AZI = AZRS
U = URS
XINC = XIRS
DMI = DMRS
E = ERS
XSINE = SINERS
FLAG2 = 0
IL = 0
IR = 0
FLAG1 = 0
NCT = 0
Is IFLAG4 = 3
No
Go To 10001
Yes
Call Subroutine SAO to Write Message to SAO on Tape 6
Write Total Flash Sequences and Total Loads on Tape 5
Call Subroutine Flex to Edit Tape 6 and Write TTY Messages on Tape 5
END

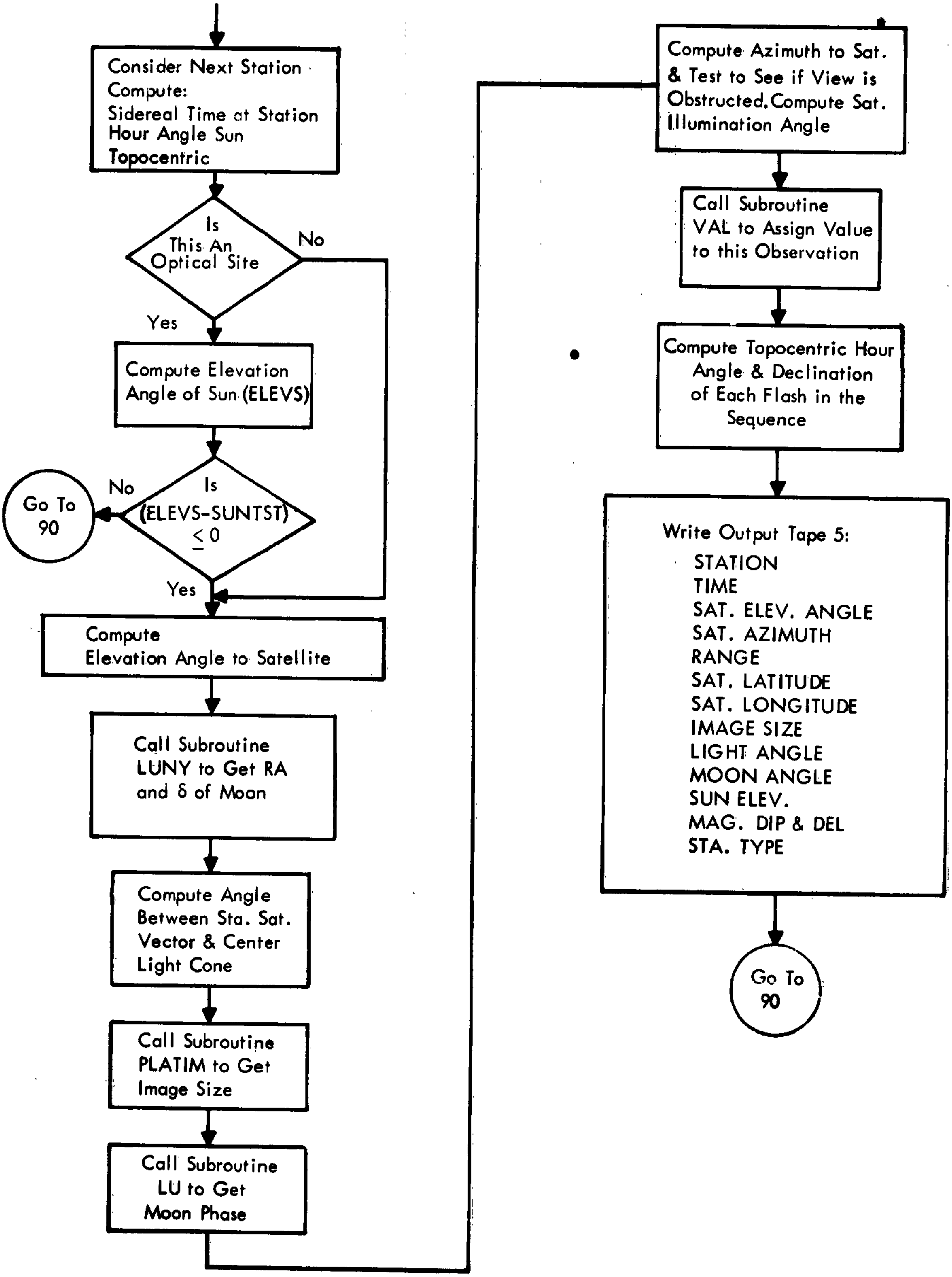

90
Have All Stations Been Considered for This Time
Is NCT - NSTA = 0
No
Go To 77
Yes
NCT = 0
91
Is OLDSIN < 1
No
Go To 80
Yes
Is SINUZ < 1
Yes
Go To 80
No
E = E - 2π
U = U - 2π
Compute New Perigee Height and Call Subroutine TLU for New Atmospheric Density and Scale Height


NIR = IR
Is NIR ≤ 0
Go To 425
No
Is (IR - IRM) ≤ 0
Yes
No
TI = 999999999.
DO 206 L = 1, NIR
Is TR (L) < 0
Yes
No
Is ((UR(L)-TI) < 0
Yes
TI = VR(L)
LL = L
206
CONTINUE
TR (LL) = -1
Is (IL+IR - ILM) ≤ 0
No
Go To 214
Yes
Do 211 L = 1, NIR
Is TR (L) < 0
Yes
IL = IL + 1
TL(IL) = TR(L)
SL(IL) = SR(L)
VL(IL) = VR(L)
211
CONTINUE
425
SET RESTART ELEMENTS
IR = 0
ILP = IL
ARS = OARS
RARS = ORARS
ECRS = OECRS
TRS = OTRS
RCRAR = ORCRAR
SINERS = OSINERS
HRS = OHRS
AZRS = OAZRS
XIRS = OXIRS
WRS = OWRS
URS = OURS
VRS = OURS
MRS = IMAS
DMRS = ODMRS
ERS = OERS


319
M = M + 1
Have All Revolutions Been Made
Is (NREV - M)<0
No
Yes
Set IFLAG4 = 3
Write on Output Tape
5 Restart Elements
Go To 273
80
Is TIME - TSTOP<0
No
Yes
Is IFIRST = 1
No
Yes
TIME = TIME + DT
STIME = TIME
Is (TIME - ELTIM)<0
No
Yes
IELTI = 2
CRAR = CRAR + GDT
TIME = STIME +DDT*CRAR
DT = DDT *GDT
CRAR = CRAR + GDT
TIME = STIME + CRAR*DDT
DT = DDT *GDT


Compute Total Perturbative Effect Of Bulge and Drag on the Elements at This Time
10001
Set Flags Telling That a New Time Has Been Computed
TNEW1 = 0
TNEW3 = 0
TNEW4 = 0
OLDSIN = sin (u_o)
SINUZ = sin ($u - u_o$)
Is IELTI = 1
Yes
Go To 80
No
Is IFRIST = 1
Yes
Go To 91
NO
COMPUTE:
Greenwich Sidereal Time
Longitude of Sun
Right Asc. of Sun
Dec. of Sun
Satellite Lctitude
Satellite Longitude
Distance from Center of Earth to Sat.


Call Subroutine AMESS To Get Magnetic Dip and Dec. of Sat.
Is DIP<0
Yes
Flash South Light
No
Flash North Light
Is This North Flash
No
Is This Time On An Odd Clock Pulse
No
Go To 91
Yes
Yes
Is This Time On An Even Clock Pulse
No
Go To 91
Yes
Do 40432 J=1, NSTA
IONE (J)=ITWO (J)
ITWO (J)=0
40432
CONTINUE


Consider Next Station
Compute:
Sidereal Time at Station
Hour Angle Sun
Topocentric
Is This An Optical Site
No
Yes
Compute Elevation Angle of Sun (ELEVS)
Is (ELEVS-SUNTST) ≤ 0
No
Go To 90
Yes
Compute Elevation Angle to Satellite
Call Subroutine LUNY to Get RA and δ of Moon
Compute Angle Between Sta. Sat. Vector & Center Light Cone
Call Subroutine PLATIM to Get Image Size
Call Subroutine LU to Get Moon Phase
Compute Azimuth to Sat. & Test to See if View is Obstructed. Compute Sat. Illumination Angle
Call Subroutine VAL to Assign Value to this Observation
Compute Topocentric Hour Angle & Declination of Each Flash in the Sequence
Write Output Tape 5:
STATION
TIME
SAT. ELEV. ANGLE
SAT. AZIMUTH
RANGE
SAT. LATITUDE
SAT. LONGITUDE
IMAGE SIZE
LIGHT ANGLE
MOON ANGLE
SUN ELEV.
MAG. DIP & DEL
STA. TYPE
Go To 90

APPENDIX I

Three teletype messages are prepared directly from magnetic tape.

Number One is to the Applied Physics Laboratory in Silver Springs, Maryland. It contains the following information:

1. Day number since January 0
2. Year number since 1900
3. Number of last injection into satellite memory + 1
4. The proper light to flash at any scheduled time
5. Time of the first flash in each sequence in milliseconds.

The format of the message is as follows:

$$\text{ANNA IB OPTICAL --- }\underbrace{\text{XXX}}_{a}\text{ --- }\underbrace{\text{XX}}_{b}\text{ - }\underbrace{\text{XXXXXX}}_{c}$$

$$\underbrace{\text{X}}_{d}\underbrace{\text{XXXXXXXXX}}_{e}$$

where a - days

b - year

c - injection number

The times given are in milliseconds from 0 hours of the day noted in the message.

A zero in d denotes a North flash and a 1 denotes a South flash.

Number Two is to NASA Goddard Space Flight Center, Greenbelt, Maryland. Contents and format are as follows:

$$\overbrace{\text{XXXXX}}^{f}\ /\ \overbrace{\text{XXXXXX}}^{g}\text{ AFCRL }\overbrace{\text{X}}^{b}\text{2 PREDICTIONS}$$

$$\underbrace{\text{XXXX}}_{i}\ /\ \underbrace{\text{XXXXX}}_{j}\ /\ \underbrace{\text{XXX}}_{k}\ /\ \underbrace{\text{XX}}_{l}\,\underbrace{\text{XX}}_{m}\,\underbrace{\text{XXX}}_{n}\,\underbrace{\text{X}}_{o}\,\underbrace{\text{XXXX}}_{p}\,\underbrace{\text{X}}_{q}\,\underbrace{\text{XXX}}_{r}$$

where

f	=	satellite identification number
g	=	year, day and month
h	=	type of look angle (R for refined, P for long range)
i	=	station number
j	=	sequence number this day
k	=	day number since Jan. 0
l	=	time of day: hour
m	=	time of day: min
n	=	time of day: seconds to nearest 1/10 second
o	=	sign of Local Hour Angle (1 indicates negative, 0 indicates positive)
p	=	Local Hour Angle to nearest 1/10 degree
q	=	sign of declination (1 indicates negative, 0 indicates positive)
r	=	declination to nearest 1/10 degree

Number Three is to the 1381st Missile Survey Squadron, Orlando, Florida. Contents and format as follows:

STA. NO.	DAY	HR	MIN	SEC	PASS NO.	SEQ. NO.	AZ(0°N)	ELEV (day)
XXXX	XXX	XX	XX	XX. XXX	XXXXXX	XXXXX	XXX. XX	XXX. XX

REFERENCES

1. Koelle, H. H. (1961); Handbook of Astronautical Engineering; McGraw-Hill

2. Baker and Makemson (1960); An Introduction to Astrodynamics; Academic Press

3. Sterne, T. S. (1959); Effect of the Rotation of a Planetary Atmosphere Upon a Close Earth Satellite; ARS Journal, October, 1959.

4. Nicolet, M. (1962); A Representation of the Terrestrial Atmosphere From 100 KM to 3000 KM; AFCRL Scientific Report No. 155.

5. Brown, D. C. (1962); Determination of Expected Image Diameters; unpublished report

6. Shea, M. A, (May 1962); Geomagnetic Field Calculations; Physics Department, University of New Hampshire, Durham, N. H.

APPENDIX II

Geomagnetic Field Calculations For a 5 Degree Polynomial

If one assumes that the earth's magnetic field at the point r, θ, and λ (radial distance, co-latitude, and east longitude) arises from purely internal sources, then the components of the field (X, Y, Z) are given by:

$$X = \sum_{n=1}^{n} \sum_{m=0}^{n} \frac{d}{d\theta} P_n^m \left[g_n^m \cos m\phi + h_n^m \sin m\phi \right] \left(\frac{R_e}{r} \right)^{n+2}$$

$$Y = -\frac{1}{\sin\theta} \left[\sum_{n=1}^{n} \sum_{m=0}^{n} P_n^m \left[-m g_n^m \sin m\phi + m h_n^m \cos m\phi \right] \left(\frac{R_e}{r} \right)^{n+2} \right]$$

$$Z = \sum_{n=1}^{n} \sum_{m=0}^{n} P_n^m \left[-(n+1)\, g_n^m \cos m\phi - (n+1)\, h_n^m \sin m\phi \right] \left(\frac{R_e}{r} \right)^{n+2}$$

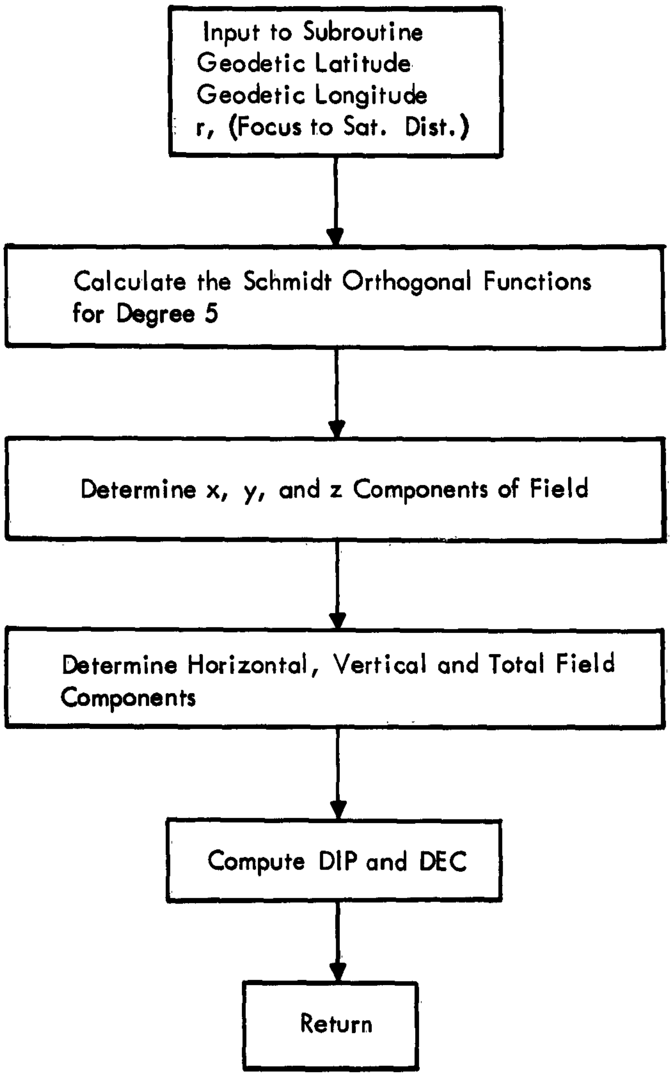
where P_n^m are the Schmidt orthogonal functions (using the associated Legendre functions; see Chapman and Bartels, Geomagnetism - Vol. II; pages 606-615); and g_n^m and h_n^m are the Schmidt coefficients of the geomagnetic field; R_e is the earth's radius in kilometers.

The total magnetic field, F, is then given by:

$$F = (X^2 + Y^2 + Z^2)^{1/2}$$

where F is calculated in gauss. To convert to gamma, the relationship 1 gauss = 10^5 gamma is used.

Flow Chart for Geomagnetic Field Computations



AF Cambridge Research Laboratories, Bedford, Mass., Geophysical Research Directorate
OPTICAL GENERATOR PROGRAM, by H.R. Kahler, R.M. Moroney, W.T. Nixon, February 1963, 40pp.
AFCRL - 63 - 445 Unclassified Report

Contains an analysis and description of a computer program written for the Philco 2000. The program computes acquisition data to a magnetically stabilized satellite that carries a flashing light on each pole. The satellite motion is described by an osculating ellipse and consideration is given to the orientation of the light, whether or not the observer is in darkness, the expected image size of the strobe light on a photographic plate, and the relative position of the moon. In addition the program selects the most geometrically "valuable" observation to make and automatically prepares three teletype messages that are sent to the observing sites.

UNCLASSIFIED

1. Astronomy, Geophyics, and Geography
2. Mathematics
3. Navigation

I. Kahler, H.R., Nixon, W.T., Moroney, R.M.

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