UNCLASSIFIED

AD NUMBER

AD385671

CLASSIFICATION CHANGES

TO: unclassified

FROM: confidential

LIMITATION CHANGES

TO:
Approved for public release, distribution unlimited

FROM:
Distribution authorized to U.S. Gov’t. agencies and their contractors; Critical technology; Oct 1967. Other requests shall be referred to Air Force Rocket Propulsion Laboratory, Edwards AFB, CA 93523.

AUTHORITY

AFRPL ltr, 18 Jan 1978; AFRPL ltr, 18 Jan 1978

THIS PAGE IS UNCLASSIFIED
SECURITY MARKING

The classified or limited status of this report applies to each page, unless otherwise marked. Separate page printouts MUST be marked accordingly.

THIS DOCUMENT CONTAINS INFORMATION AFFECTING THE NATIONAL DEFENSE OF THE UNITED STATES WITHIN THE MEANING OF THE ESPIONAGE LAWS, TITLE 18, U.S.C., SECTIONS 793 AND 794. THE TRANSMISSION OR THE REVELATION OF ITS CONTENTS IN ANY MANNER TO AN UNAUTHORIZED PERSON IS PROHIBITED BY LAW.

NOTICE: When government or other drawings, specifications or other data are used for any purpose other than in connection with a definitely related government procurement operation, the U. S. Government thereby incurs no responsibility, nor any obligation whatsoever; and the fact that the Government may have formulated, furnished, or in any way supplied the said drawings, specifications, or other data is not to be regarded by implication or otherwise as in any manner licensing the holder or any other person or corporation, or conveying any rights or permission to manufacture, use or sell any patented invention that may in any way be related thereto.
CONFIDENTIAL

AFRPL-TR-67-274

(Unclassified Title)

ENGINEERING PROPERTY DATA ON ROCKET PROPELLANTS

Fifth Quarterly Report

Rocketdyne
A Division of North American Rockwell Corporation
6633 Canoga Avenue
Canoga Park, California

October 1967

Group 4
Downgraded at 3-Year Intervals
Declassified After 12 Years

THIS MATERIAL CONTAINS INFORMATION AFFECTING THE NATIONAL DEFENSE OF THE UNITED STATES WITHIN THE MEANING OF THE ESPIONAGE LAWS, TITLE 18 U.S.C. SECTIONS 793 AND 794. THE TRANSMISSION OR REVELATION OF WHICH IN ANY MANNER TO AN UNAUTHORIZED PERSON IS PROHIBITED BY LAW.

In addition to security requirements which must be met, this document is subject to special export controls and each transmittal to foreign governments or foreign nationals may be made only with prior approval of AFRPL (RPPR-STINFO), Edwards, California 93523.

Air Force Rocket Propulsion Laboratory
Research and Technology Division
Edwards Air Force Base, California
Air Force Systems Command
United States Air Force
When U.S. Government drawings, specifications, or other data are used for any purpose other than a definitely related Government procurement operation, the Government thereby incurs no responsibility nor any obligation whatsoever, and the fact that the Government may have formulated, furnished, or in any way supplied the said drawings, specifications, or other data, is not to be regarded by implication or otherwise, or in any manner licensing the holder or any other person or corporation, or conveying any rights or permission to manufacture, use, or sell any patented invention that may in any way be related thereto.
NOTICE
UNITED STATES PATENT OFFICE

A patent application has been filed in the U.S. Patent Office by North American Aviation, Inc., based upon subject matter included herein or related hereto, and the Secrecy Order appended hereto has been issued thereon pursuant to Title 35, United States Code (1952) Sections 181-188. Further dissemination of said subject matter is prohibited except in strict compliance with said order. The recipient of this document is requested to notify all persons who will have access to this material of the Secrecy Order. Penalties for violation of a Secrecy Order include a fine of up to $10,000 or imprisonment for not more than two years, or both.

DEPARTMENT OF COMMERCE
United States Patent Office
Washington

SECRET ORDER

NOTE: To the applicant above named, his heirs, and any and all his assignees, attorneys and agents, hereafter designated principals:

You are hereby notified that your application as above identified has been found to contain subject matter, the unauthorized disclosure of which might be detrimental to the public safety or defense, and you are ordered in writing to publish or disclose the invention or any material information with respect hereto, including all to unpublished details of the subject matter of said application, to any person not cognizant of the invention prior to the date of the order, including any employee of the principals, but to keep the same secret except by written permission from the Commissioner of Patents, under the penalties of 35 U.S.C. (1952) 183, 186.

Any other application which contains any significant part of the subject matter of the above identified application falls within the scope of this order. If such other application does not stand under a secrecy order, the common subject matter shall be brought to the attention of the Patent Security Division, Patent Office.

If prior to the issuance of the secrecy order any significant part of the subject matter has been revealed to any person, the principals shall promptly inform such person of the secrecy order and the penalties for improper disclosure.

This order should not be construed in any way to mean that the Government has adopted or contemplates adoption of the alleged invention disclosed in this application; nor is it any indication of the value of such invention.

PERMIT A

An order of secrecy having been issued in the above-entitled application by the Commissioner of Patents, the principals as designated in said order are authorized to disclose the subject matter to any person of the classes hereinafter specified if such person is known to the principals disclosing to be concerned directly in an official capacity with the subject matter, providing that all reasonable safeguards are taken to otherwise protect the invention from unauthorized disclosure. The specified classes are:

(a) Any officer or employee of any department, independent agency or bureau of the Government of the United States.
(b) Any person designated specifically by the head of any department, independent agency or bureau of the Government of the United States, or by his duly authorized subordinate, as a proper individual to receive the disclosure of the above indicated application.

The principals under the secrecy order are further authorized to disclose the subject matter of this application to the minimum necessary number of persons of known loyalty and discretion, employed by or working with the principals or their licensees and whose duties involve cooperation in the development, manufacture or use of the subject matter by or for the Government of the United States, provided such persons are advised of the issuance of the secrecy order.

The provisions of this permit do not in any way lessen responsibility for the security of the subject matter as imposed by any Government contract of the provisions of the existing laws relating to espionage and national security.

First Assistant Commissioner
ENGINEERING PROPERTY DATA ON ROCKET PROPELLANTS
Fifth Quarterly Report

Rocketdyne
A Division of North American Rockwell Corporation
Canoga Park, California

October 1967

In addition to security requirements which must be met, this document is subject to special export controls and each transmittal to foreign governments or foreign nationals may be made only with prior approval of AFRPL (RPDR-STINFO), Edwards, California 93523.

Air Force Rocket Propulsion Laboratory
Research and Technology Division
Edwards Air Force Base, California
Air Force Systems Command
United States Air Force
FOREWORD

(U) This is a quarterly technical report submitted in compliance with Contract AF04(611)-11407. The research reported herein, which covers the period 1 July through 30 September 1967, was sponsored by the Air Force Rocket Propulsion Laboratory, Research and Technology Division, Air Force Systems Command, United States Air Force, Edwards, California, with Mr. G. Allen Beale acting as the Air Force Project Officer.

(U) This program was conducted in the Chemical and Material Sciences Department of the Rocketdyne Research Division, with Dr. E. F. C. Cain serving as Program Manager and Mr. M. T. Constantine serving as the Responsible Project Scientist. Other technical personnel who have contributed to this effort include K. J. Youel (Phases I and III), Dr. J. F. Hon (Phase II), Dr. W. Unterberg (Phase II), Dr. S. E. Rodriguez (Phase II), J. V. Lecce (Phase II), R. W. Melvold (Phase II), and J. Quaglino (Phase II).

(U) This report has been assigned the Rocketdyne identification number R-6638-5.

(U) Publication of this report does not constitute Air Force approval of the report's findings or conclusions. It is published only for the exchange and stimulation of ideas.

W. H. Ebelke, Colonel, USAF
Chief, Propellant Division
The results of the sixth quarter year's effort of a 24-month program on the analytical and experimental characterization of the physical properties of selected liquid propellants are presented in three phases. In Phase I, a continuous review of the literature was conducted to ensure the acquisition and documentation of the latest possible propellant properties data for evaluation and possible inclusion into a propellant properties handbook. Phase II experimental efforts have resulted in additional measurements of chlorine trifluoride pressure-volume-temperature relationships in the critical region and the establishment of vapor pressure equations. Sonic velocity measurements were conducted in chlorine trifluoride and chlorine pentafluoride under pressurized conditions, and in saturated liquid nitrogen tetroxide and MIF-3. Nitrogen gas solubility was determined in chlorine pentafluoride and the viscosity of chlorine pentafluoride was measured at high pressures. The specific heats of MIF-3 and MIF-5 were measured. Phase III efforts included the evaluation and assembly of all data generated in Phases I and II and the compilation of surface tension data on chlorine trifluoride, chlorine pentafluoride, nitrogen tetroxide, 50 N_2H_4-50(CH_3)_2N_2H_2 fuel blend, hydrazine, MMH, UDMH, MIF-3, and MIF-5.
CONTENTS

Forword .................................................. ii
Abstract .................................................. iii
Introduction ............................................. 1
Summary .................................................. 3
Phase I: Literature Search ............................... 9
  Objective .............................................. 9
  Results and Accomplishments ......................... 9
Phase II: Experimental Determinations ................. 11
  Objective .............................................. 11
  Results and Accomplishments ......................... 11
Phase III: Evaluation and Compilation of Data ........ 43
  Objective .............................................. 43
  Results and Accomplishments ......................... 43
References ............................................... 53
<table>
<thead>
<tr>
<th></th>
<th>Illustrations</th>
<th>Page</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>Vapor Pressure of Chlorine Trifluoride</td>
<td>15</td>
</tr>
<tr>
<td>2</td>
<td>Sonic Velocity Apparatus</td>
<td>17</td>
</tr>
<tr>
<td>3</td>
<td>Sonic Velocity in CIF₃ and CIF₅</td>
<td>19</td>
</tr>
<tr>
<td>4</td>
<td>Velocity of Sound in Saturated Liquid Nitrogen Tetroxide</td>
<td>23</td>
</tr>
<tr>
<td>5</td>
<td>Adiabatic Compressibility of Saturated Liquid Nitrogen Tetroxide</td>
<td>24</td>
</tr>
<tr>
<td>6</td>
<td>Differential Solubility of Gaseous Nitrogen in Chlorine Pentfluoride</td>
<td>28</td>
</tr>
<tr>
<td>7</td>
<td>Total Solubility of Gaseous Nitrogen in Chlorine Pentfluoride</td>
<td>30</td>
</tr>
<tr>
<td>8</td>
<td>Specific Heat of MHF-3</td>
<td>33</td>
</tr>
<tr>
<td>9</td>
<td>Specific Heat of MHF-5</td>
<td>37</td>
</tr>
<tr>
<td>10</td>
<td>Viscosity of Saturated Liquid Chlorine Pentfluoride</td>
<td>40</td>
</tr>
<tr>
<td>11</td>
<td>Surface Tension of Saturated Liquid Chlorine Trifluoride</td>
<td>45</td>
</tr>
<tr>
<td>12</td>
<td>Surface Tension of Saturated Liquid Chlorine Pentfluoride</td>
<td>46</td>
</tr>
<tr>
<td>13</td>
<td>Surface Tension of Anhydrous Hydrazine</td>
<td>47</td>
</tr>
<tr>
<td>14</td>
<td>Surface Tension of Monomethylhydrazine</td>
<td>49</td>
</tr>
<tr>
<td>15</td>
<td>Surface Tension of UDMH</td>
<td>50</td>
</tr>
</tbody>
</table>
TABLES

1. Chlorine Trifluoride Vapor Pressure Data .................................................. 14
2. Velocity of Sound in Saturated Liquid Chlorine Trifluoride .................................................. 20
3. Velocity of Sound in Saturated Liquid Nitrogen Tetroxide .................................................. 22
4. Nitrogen Gas Solubility in Chlorine Pentafluoride .................................................. 27
5. Experimental Specific Heat of MUF-3 Fuel Blend .................................................. 32
6. Experimental Specific Heat of MHF-5 Fuel Blend .................................................. 35
7. Experimental Chlorine Pentafluoride Viscosity Data .................................................. 39
INTRODUCTION

(U) Under Contract AF04(611)-10546, Rocketdyne completed a 12-month analytical and experimental program (Ref. 1) on the rational and systematic physical characterization of selected liquid rocket propellants. The overall objective of this program, which was conceived as an initial step toward elimination of propellant data gaps, was to assemble and experimentally complete the data on essential physical properties of current and near-term liquid rocket propellants over temperature and pressure ranges of practical use to propulsion engineering. During this program, experimental efforts resulted in measurement of: (1) chlorine trifluoride (CIF₃) phase properties (including critical temperature); (2) IRFNA and chlorine pentafluoride (CIF₅) sonic velocities (and calculation of compressibilities); (3) CIF₃ and CH₃N₂H₃ specific heats (and correction of previously determined CIF₅ specific heat data); (4) thermal conductivities of the 50 N₂H₄ - 50 (CH₃)₂N₂H₂ fuel blend and CH₃N₂H₃; and (5) the design and preliminary assembly of apparatus for measurement of inert-gas solubility in liquids and liquid viscosities at extended temperatures and pressures. Analytical efforts, initiated with an extensive literature survey, included the assembly and evaluation of physical property data on MHF-1, MHF-3, MHF-5, CIF₃, and CIF₅ for future correlation and summary publication.

(U) The present three-phase program being conducted under Contract AF04(611)-11407, represents a 24-month extension and expansion of the objectives of the previous effort. Phase I effort consists of a continuous review of the current literature to document reported propellant properties. In Phase II, effort is directed at the experimental determination of unavailable engineering data, for selected oxidizers and fuels, which are required to design rocket engine hardware. Effort under Phase III includes the compilation, correlation, and evaluation of all data obtained from Phases I and II and presentation of the valid data in an annual technical report.

(U) During the first 15 months of the current program, which has been reported in an Annual Summary Report (Ref. 2) and in a subsequent Quarterly
Report (Ref. 3), the continuous documentation of current propellant properties data was maintained with a detailed review of 579 reports and of all current releases of the standard abstract sources. Experimental efforts in this 15-month period resulted in measurements of: (1) CIF$_3$ density and vapor pressure over a wide temperature range (with a corresponding estimation of critical properties); (2) saturated and pressurized liquid CIF$_5$ and CIF$_3$ sonic velocities (and calculation of adiabatic compressibilities); (3) N$_2$(g) solubility in CIF$_5$, (4) UDMH, 50 N$_2$H$_4$-50 (CH$_3$)$_2$N$_2$H$_2$, CIF$_3$, and WHF-3 specific heats; (5) thermal conductivities of UDMH, WHF-3, and MHF-5; and (6) CIF$_3$ viscosity. Phase III analytical efforts resulted in correlation of Phase II experimental data; the preparation of complete physical property bibliographies on B$_2$H$_6$ and N$_2$H$_4$ and the initiation of N$_2$O$_4$ and MMH physical property bibliographies; the publication of N$_2$H$_4$ physical property data sheets; a curve fit of specific heat data on solid CIF$_3$; and a report of unpublished data on CIF$_5$ boiling point and CIF$_5$ storability.

(U) Progress achieved in a continuation of these objectives and efforts during the sixth quarter year of the current program is summarized in this report.
SUMMARY

(U) Analytical and experimental research conducted during the sixth quarter year of a current 24-month program to complete the data on essential physical properties of current and near-term liquid propellants of interest to the Air Force is described in three phases.

(U) In Phase I, a continuous survey of the current propellant literature has resulted in the preliminary screening of 1995 reports acquired and cataloged by Rocketdyne during this period; 162 of these reports were reviewed in detail for pertinent engineering properties data on liquid propellants.

(U) The experimental characterization of essential physical properties of selected propellants was conducted under Phase II. During the current report period, these efforts were directed at measurements of: CINF3 phase properties in the critical region; sonic velocity in CINF5 and CINF3 under pressurized conditions; sonic velocity in saturated liquid N2O4 and MIF-3; gaseous nitrogen solubility in CINF5; specific heat of MIF-3 and MIF-5; and CINF5 viscosity.

(U) A constant-volume vapor pressure apparatus was used to extend the available experimental data on CINF3 phase properties. As a result of the current experimental data, the vapor pressure of CINF3 was correlated from -30 to 179 C (-22 to 394 F) with the following equations:

\[
\log P_{(\text{atm})} = 4.5064 - \frac{1101.65}{t(\text{C}) + 233.0}
\]

and

\[
\log P_{(\text{psia})} = 5.6733 - \frac{1982.97}{t(\text{F}) + 387.4}
\]
Data resulting from measurements of sonic velocity in ClF₅ and ClF₃ under total pressurizations of 500 and 1000 psia were correlated and curve fit with the following equations:

**500 psia**

ClF₅: \( c(\text{m/sec}) = 1696 - 3.642T(K) \)

and

\( c(\text{ft/sec}) = 5564.3 - 6.637T(R) \)

ClF₃: \( c(\text{m/sec}) = 2007 - 3.878T(K) \)

and

\( c(\text{ft/sec}) = 6584.6 - 7.067T(R) \)

**1000 psia**

ClF₅: \( c(\text{m/sec}) = 1670 - 3.487T(K) \)

and

\( c(\text{ft/sec}) = 5479.0 - 6.355T(R) \)

ClF₃: \( c(\text{m/sec}) = 2008 - 3.847T(K) \)

and

\( c(\text{ft/sec}) = 6587.9 - 6.995T(R) \)
Results from experimental determinations of sonic velocity in saturated liquid N\textsubscript{2}O\textsubscript{4} over a temperature range of -5 to 56 C (23 to 132 F) were curve fit with the following equations:

\[ c(\text{m/sec}) = 2296 - 4.425 T(K) \]

and

\[ c(\text{ft/sec}) = 3825 - 8.065 t(F) \]

Adiabatic compressibilities calculated from these sonic velocity data and corresponding temperature-N\textsubscript{2}O\textsubscript{4} density data established the following relationships:

\[ \beta_{\text{atm}^{-1}} = 5.776 \times 10^{-5} + 5.454 \times 10^{-7} t(\text{C}) + 5.185 \times 10^{-9} t(\text{C})^2 \]

\[ - 1.659 \times 10^{-11} t(\text{C})^3 + 5.291 \times 10^{-13} t(\text{C})^4 \]

and

\[ \beta_{\text{psi}^{-1}} = 3.394 \times 10^{-6} + 1.260 \times 10^{-8} t(F) + 1.486 \times 10^{-11} t(F)^2 \]

\[ - 6.329 \times 10^{-13} t(F)^3 + 3.431 \times 10^{-15} t(F)^4 \]

Preliminary measurements of sonic velocity in MIF-3 resulted in a value of 1604 m/sec (5262 ft/sec) ±4 percent at 25 C (77 F).

Measurements of gaseous nitrogen solubility in liquid ClF\textsubscript{3} were extended to higher temperature and pressure levels. As a result, the following expressions have been derived for total solubility (S, lb N\textsubscript{2}/lb ClF\textsubscript{3}) as a function of pressure (P, psia) at temperatures of 90, 120, 150, and 180 F, respectively:

\[ S = 1.62 \times 10^{-5} (P-72) + 0.36 \times 10^{-8} (P-72)^2 \]
S = 1.97 \times 10^{-5} (P-115) + 0.33 \times 10^{-8} (P-115)^2

S = 2.30 \times 10^{-5} (P-176) + 0.31 \times 10^{-8} (P-176)^2

S = 2.80 \times 10^{-5} (P-257)

(C) Specific heat measurements on liquid MHF-3 were completed over the temperature range of -56.0 to 47.3 C (-50.8 to 117.2 F) through the use of an adiabatic calorimeter. A curve fit of the experimental data resulted in the following equations:

\[ C_p(\text{cal/gm-C}) = 0.708 + 3.807 \times 10^{-4} t(C) + 2.110 \times 10^{-6} t(C)^2 \]

and

\[ C_p(\text{Btu/lb-F}) = 0.702 + 1.698 \times 10^{-4} t(F) + 6.512 \times 10^{-7} t(F)^2 \]

(C) Experimental specific heat data obtained on liquid MHF-5 over a temperature range of -47.4 to 7.5 C (-53.3 to 45.5 F) in this program were correlated with other available MHF-5 specific heat data at higher temperatures. This correlation resulted in the following MHF-5 specific heat-temperature relationships over a temperature range of -47.4 to 68.0 C (-53.3 to 154.4 F):

\[ C_p(\text{cal/gm-C}) = 0.664 + 1.21 \times 10^{-4} t(C) \]

and

\[ C_p(\text{Btu/lb-F}) = 0.662 + 6.74 \times 10^{-5} t(F) \]

(U) The applicability of the present steady-state, concentric-cylinder conductivity cell to oxidizer thermal conductivity measurements is currently being evaluated. Upon assurance of a suitable and compatible apparatus and technique, thermal conductivity measurements will be initiated on CIF5.
Liquid CI\textsubscript{5} viscosity measurements at 202 F in a stainless-steel capillary viscometer over a total pressure range of 360 to 910 psia indicated that these pressures have no significant effect on the CI\textsubscript{5} kinematic viscosity. The absolute viscosity of the saturated liquid at this temperature is 0.168 centipoise.

Efforts under the Phase III review and evaluation of the data generated under Phases I and II were directed at the continuation of the assembly of engineering property bibliographies and physical property data sheets on N\textsubscript{2}O\textsubscript{4}. To support specific Phase II requirements, experimental N\textsubscript{2}O\textsubscript{4} density data were assembled and correlated over a temperature range of -11.2 to 95.3 C (11.8 to 199.9 F). In addition, all published surface tension data on CI\textsubscript{5}, CI\textsubscript{5}, N\textsubscript{2}O\textsubscript{4}, 50 N\textsubscript{2}H\textsubscript{4}-50 UDMH, N\textsubscript{2}H\textsubscript{4}, NOH, UDMH, MIF-3, and MIF-5 were compiled and summarized.
PHASE I: LITERATURE SEARCH

OBJECTIVE

(U) The Phase I objective is the maintenance of a continuous review of the current literature and efforts of other investigators in the field to ensure acquisition and documentation of the latest possible propellant properties data for evaluation and possible inclusion into a propellant properties handbook. This survey is designed to include, but not to be necessarily limited to, the properties of the following fuels and oxidizers:

<table>
<thead>
<tr>
<th>Oxidizers</th>
<th>Fuels</th>
</tr>
</thead>
<tbody>
<tr>
<td>Liquid Oxygen (LO₂)</td>
<td>Liquid Hydrogen (LH₂)</td>
</tr>
<tr>
<td>Chlorine Pentafluoride (ClF₅)</td>
<td>50 N₂H₄ - 50 (CH₃)₂N₂H₂ Fuel Blend</td>
</tr>
<tr>
<td>Chlorine Trifluoride (ClF₃)</td>
<td>Hydrazine (N₂H₄)</td>
</tr>
<tr>
<td>Fluorine (F₂)</td>
<td>MMH (CH₃N₂H₃)</td>
</tr>
<tr>
<td>Hydrogen Peroxide (H₂O₂)</td>
<td>UDMH [(CH₃)₂N₂H₂]</td>
</tr>
<tr>
<td>Nitrogen Tetroxide (N₂O₄)</td>
<td>N₂H₄-MMH Mixtures</td>
</tr>
<tr>
<td>Mixed Oxides of Nitrogen (N₂O₄-NO)</td>
<td>Hybaline B-3</td>
</tr>
<tr>
<td>FLOX Mixtures (LF₂-LO₂)</td>
<td>Alumizine</td>
</tr>
<tr>
<td>Oxygen Difluoride (OF₂)</td>
<td>Pentaborane (B₃H₉)</td>
</tr>
<tr>
<td>Tetrafluorohydrazine (N₂F₄)</td>
<td>Diborane (B₂H₆)</td>
</tr>
<tr>
<td></td>
<td>MHF Blends</td>
</tr>
<tr>
<td></td>
<td>MAF Blends</td>
</tr>
</tbody>
</table>

RESULTS AND ACCOMPLISHMENTS

(U) A formal survey of current propellant literature, initiated under Contract AF04(611)-10546 (Ref. 1), is being continued as Phase I of the present program. This survey, which includes the location, acquisition,
and documentation of all available propellant properties data of interest to the Air Force, was originally directed at a comprehensive review of physical properties data. However, under the present contract, the survey has been expanded to include additional engineering properties data.

(U) This survey is being maintained through two primary techniques. The main portion of the effort is directed at the survey of all reports acquired by Rocketdyne; these include reports obtained as a result of individual personnel requests as well as those acquired through normal distribution channels. Each of these reports is surveyed with respect to subject matter, and reports containing potential propellant properties data are selected for detailed review. All pertinent data contained in these reports are documented for assembly under Phase III.

(U) To aid in the complete awareness of all available propellant properties data and their subsequent acquisition and documentation, this report survey effort is supplemented by a continuous survey of the current releases of Chemical Abstracts, NASA CSTAR Abstracts, CPIA Chemical Propulsion Abstracts, Defense Documentation Center (DDC) Technical Abstract Bulletin (TAB), NBS Cryogenic Data Center Current Awareness Service, and propellant manufacturers' bulletins. Any pertinent reports located through these sources that have not been acquired previously by Rocketdyne are ordered and reviewed in detail.

(U) During the current report period, 1995 reports were surveyed; of this total, 162 reports were reviewed in detail for propellant properties data. This current review included report bibliographies on CH\textsubscript{3}N\textsubscript{2}H\textsubscript{3} and N\textsubscript{2}O\textsubscript{4}, which were obtained (by Rocketdyne request) from the Defense Documentation Center. The pertinent data abstracted from these reports are being compiled and evaluated under Phase III.
PHASE II: EXPERIMENTAL DETERMINATIONS

OBJECTIVE

(U) Phase II is designed for the experimental characterization of essential physical properties of selected liquid propellants. This phase essentially constitutes a 24-month continuation of the efforts initiated under Phase II of Contract AF04(611)-10546 (Ref. 1). Selection of the propellants and properties to be characterized experimentally is related to the unavailability of required data and relative importance of the data to the Air Force. Initial efforts have emphasized the completion of those propellant properties recommended for initial characterization under the previous program. Additional efforts are continuing in an order related to the importance of the data to the Air Force as determined by the Air Force Project Engineer.

(U) The selected properties are being determined over the liquidus temperature range and over a pressure range of 14.7 to 1000 psi, where practical. Changes to the selected list can be made at any time during the program through mutual agreement of Rocketdyne and the Air Force Contracting Officer. Standard test methods are being used when available. Unique or new test methods to be used must be approved by the Air Force Contracting Officer.

RESULTS AND ACCOMPLISHMENTS

(U) Phase II efforts during the current period were directed at:

1. Continued characterization of ClF₃ phase properties in the high-temperature and critical region

2. Experimental measurement of sonic velocity in saturated N₂O₄ and MIBJ-3 and correlation of previously obtained data on sonic velocity in ClF₅ and ClF₃ at pressures above saturation
3. Continuation of \( \text{N}_2(\text{g}) \) solubility measurements in \( \text{ClF}_5 \)
4. Completion of MHF-3 and MHF-5 specific heat measurements
5. Initiation of thermal conductivity measurements on selected oxidizers
6. Extension of the \( \text{ClF}_5 \) viscosity measurements

The efforts conducted under each of these tasks are described in the following paragraphs.

**Phase Property Measurements**

(U) Current phase property measurements have been designed for the reliable extension of the available experimental data on \( \text{ClF}_3 \) into the critical range. Previous efforts (Ref. 2) in this program had expanded and correlated available density data (Ref. 1 and 4) on saturated liquid \( \text{ClF}_3 \) over a temperature range of -22 to 161°C (-8 to 322°F). Vapor pressure measurements (Ref. 2 and 3) on \( \text{ClF}_3 \) had been conducted over a temperature range of 25.65 to 148.58°C (78.17 to 299.4°F) to increase the reliability of previously available (Ref. 1 and 5) vapor pressure data in the high-temperature region. In addition, a \( \text{ClF}_3 \) critical temperature measurement (using the liquid-vapor meniscus disappearance technique) of 179.6 ±0.5°C (355.3°F) was reported (Ref. 1) and used (Ref. 3) with the preliminary vapor pressure data to calculate a critical pressure of 961 psia (65.4 atm) and a critical density of 0.88 g/cc (54.9 lb/cu ft).

(U) The expansion of these phase property measurements in the critical region is being accomplished through the experimental determination of pressure-temperature relationships of known quantities of \( \text{ClF}_3 \) using a constant-volume vapor pressure apparatus (which has been described previously in Ref. 2). To ensure attainment of reliable data, the 0- to 1000-psia range pressure transducer used with the apparatus was calibrated with a Heise gage at temperatures and pressures over the intended ranges.
3. Continuation of \( \text{N}_2(g) \) solubility measurements in CI\( \text{F}_5 \)

4. Completion of MHF-3 and MHF-5 specific heat measurements

5. Initiation of thermal conductivity measurements on selected oxidizers

6. Extension of the CI\( \text{F}_5 \) viscosity measurements

The efforts conducted under each of these tasks are described in the following paragraphs.

**Phase Property Measurements**

(U) Current phase property measurements have been designed for the reliable extension of the available experimental data on CI\( \text{F}_3 \) into the critical range. Previous efforts (Ref. 2) in this program had expanded and correlated available density data (Ref. 1 and 4) on saturated liquid CI\( \text{F}_3 \) over a temperature range of -22 to 161 C (-8 to 322 F). Vapor pressure measurements (Ref. 2 and 3) on CI\( \text{F}_3 \) had been conducted over a temperature range of 25.65 to 148.58 C (78.17 to 299.4 F) to increase the reliability of previously available (Ref. 1 and 5) vapor pressure data in the high-temperature region. In addition, a CI\( \text{F}_3 \) critical temperature measurement (using the liquid-vapor meniscus disappearance technique) of 179.6 ±0.5 C (355.3 F) was reported (Ref. 1) and used (Ref. 3) with the preliminary vapor pressure data to calculate a critical pressure of 961 psia (65.4 atm) and a critical density of 0.88 gm/co (54.9 lb/cu ft).

(U) The expansion of these phase property measurements in the critical region is being accomplished through the experimental determination of pressure-temperature relationships of known quantities of CI\( \text{F}_3 \) using a constant-volume vapor pressure apparatus (which has been described previously in Ref. 2). To ensure attainment of reliable data, the 0- to 1000-psia range pressure transducer used with the apparatus was calibrated with a Heise gage at temperatures and pressures over the intended ranges.
of use. Because the calibration results showed that the transducer output was significantly sensitive to temperature changes, a second-order function was required in both the output voltage and input temperature terms during data reduction. A curve fit of the calibration data with an appropriate equation was obtained using a least-squares computer program.

(1) Chlorine Trifluoride Vapor Pressure. During the present effort, additional pressure-temperature measurements (using variations of the amount of propellant-grade ClF₃ in the apparatus) were conducted in the constant-volume, vapor-pressure apparatus over a temperature range of 4.20 to 146.60 °C (39.56 to 295.88 °F). The resulting vapor pressure data are given in Table 1. These data were combined with the previously reported values (Ref. 2, 3, and 5) and correlated with an Antoine equation using a least-squares computer program. The resulting curve fit (Fig. 1) gave the following equations over the temperature range of -30 to 179 °C (-22 to 354 °F):

\[
\log P_{\text{atm}} = 4.5064 - \frac{1101.65}{t(°C)} + 233.0
\]

and

\[
\log P_{\text{psia}} = 5.6733 - \frac{1082.97}{t(°F)} + 387.4
\]

The large (5 percent) standard deviation of these curve fits is somewhat reasonable for a 3-constant equation over the large-temperature range; however, additional measurements are indicated. Using these vapor pressure equations and the experimental critical temperature of 179.6 °C (355.3 °F), a critical pressure of 68.6 atm (1008 psia) can be calculated.
<table>
<thead>
<tr>
<th>Sample No.</th>
<th>Temperature</th>
<th>Vapor Pressure, psia</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>C</td>
<td>F</td>
</tr>
<tr>
<td>1</td>
<td>40.08</td>
<td>104.14</td>
</tr>
<tr>
<td>1</td>
<td>66.80</td>
<td>152.24</td>
</tr>
<tr>
<td>1</td>
<td>96.00</td>
<td>204.80</td>
</tr>
<tr>
<td>2</td>
<td>39.28</td>
<td>102.70</td>
</tr>
<tr>
<td>2</td>
<td>61.83</td>
<td>143.29</td>
</tr>
<tr>
<td>2</td>
<td>78.58</td>
<td>173.44</td>
</tr>
<tr>
<td>2</td>
<td>101.08</td>
<td>213.94</td>
</tr>
<tr>
<td>3</td>
<td>4.20</td>
<td>39.56</td>
</tr>
<tr>
<td>3</td>
<td>45.43</td>
<td>113.77</td>
</tr>
<tr>
<td>3</td>
<td>61.43</td>
<td>142.57</td>
</tr>
<tr>
<td>3</td>
<td>75.78</td>
<td>168.40</td>
</tr>
<tr>
<td>3</td>
<td>93.28</td>
<td>199.90</td>
</tr>
<tr>
<td>3</td>
<td>122.30</td>
<td>252.14</td>
</tr>
<tr>
<td>3</td>
<td>146.60</td>
<td>295.88</td>
</tr>
</tbody>
</table>
Figure 1. Vapor Pressure of Chlorine Trifluoride

Curves fit from -32 to 354°F with equation:

$$\log_{10}(\text{Pressure (PSIA)}) = 5.6733 + 1.903.92 \quad \left(\text{F} - 397.4\right)$$
Sonic Velocity (and Compressibility) Measurements

Determinations of sonic velocity and adiabatic compressibility of ClF₅, ClF₃, and N₂O₄ have been conducted with an apparatus described in Ref. 2. A similar apparatus (Fig. 2) is being used for sonic velocity measurements in MFF-3. With these apparatuses, the velocity of sound in a liquid is obtained experimentally by measuring the time required for an ultrasonic acoustic wave to travel a known distance in the test liquid. Using the resulting velocity of sound (c) and the density (ρ) of the liquid at the corresponding temperature, the adiabatic compressibility (βₐ) of the liquid is calculated from the relationship:

\[ \beta_a = \frac{1}{\rho c^2} \]

Sonic Velocity in Chlorine Pentfluoride. The data from previous measurements (Ref. 3) of sonic velocity in ClF₅ under pressurized (with GN₂) liquid conditions were curve fit using a least-squares computer program. The resulting equations with their standard deviation (σ) are as follows:

500-psia total pressure, -15 to 69 °C (5 to 156.2 °F)

\[ c (\text{m/sec}) = 1696 - 3.642 \ T(K) \quad (\sigma = \pm 0.7 \ \text{m/sec}) \]

and

\[ c (\text{ft/sec}) = 5564.3 - 6.637 \ T(°R) \quad (\sigma = \pm 2.3 \ \text{ft/sec}) \]

1000-psia total pressure, -15 to 46 °C (5.0 to 114.8 °F)

\[ c (\text{m/sec}) = 1670 - 3.487 \ T(K) \quad (\sigma = \pm 1.6 \ \text{m/sec}) \]

and

\[ c (\text{ft/sec}) = 5479.0 - 6.355 \ T(°R) \quad (\sigma = \pm 5.2 \ \text{ft/sec}) \]
Figure 2. Sonic Velocity Apparatus
The curve fits expressed in the metric units are graphically illustrated in Fig. 3 and compared with previously established (Ref. 2) data for the saturated liquid.

(U) Sonic Velocity in Chlorine Trifluoride. During this report period, sonic measurements were completed in liquid ClF₃ under total pressurizations (with gaseous nitrogen) of 500, 800, and 1000 psia using the experimental procedure previously described in Ref. 3. The resulting data are shown in Table 2 with the previously established (Ref. 2) saturated liquid data. The data from the measurements at 500 and 1000 psia were curve fit (through a least-squares computer program) with the following equations:

**500-psia total pressure, -14 to 75°C (6.8 to 167.0°F)**

\[
c(\text{m/sec}) = 2007 - 3.878 T(\text{K}) \quad (\sigma = \pm2.2 \text{ m/sec})
\]

and

\[
c(\text{ft/sec}) = 6584.6 - 7.067 T(\text{R}) \quad (\sigma = \pm7.2 \text{ ft/sec})
\]

**1000-psia total pressure, -14 to 56°C (6.8 to 132.8°F)**

\[
c(\text{m/sec}) = 2008 - 3.837 T(\text{K}) \quad (\sigma = \pm1.5 \text{ m/sec})
\]

and

\[
c(\text{ft/sec}) = 6587.9 - 6.995 T(\text{R}) \quad (\sigma = \pm4.9 \text{ ft/sec})
\]

Plots of these and the saturated liquid data curve fits are illustrated (in metric units) in Fig. 3.
Figure 3. Sonic Velocity in ClF₃ and ClF₅.
TABLE 2

VELOCITY OF SOUND IN SATURATED LIQUID CHLORINE TRIFLUORIDE

<table>
<thead>
<tr>
<th>Temperature</th>
<th>Total Pressure, psia</th>
<th>Sonic Velocity</th>
</tr>
</thead>
<tbody>
<tr>
<td>C</td>
<td>F</td>
<td>m/sec</td>
</tr>
<tr>
<td>-14.12</td>
<td>6.6</td>
<td></td>
</tr>
<tr>
<td></td>
<td>Saturation*</td>
<td>991.9</td>
</tr>
<tr>
<td></td>
<td>500</td>
<td>1002.6</td>
</tr>
<tr>
<td></td>
<td>800</td>
<td>1010.0</td>
</tr>
<tr>
<td></td>
<td>1000</td>
<td>1013.5</td>
</tr>
<tr>
<td>4.43</td>
<td>40.0</td>
<td></td>
</tr>
<tr>
<td></td>
<td>Saturation*</td>
<td>923.1</td>
</tr>
<tr>
<td></td>
<td>500</td>
<td>933.5</td>
</tr>
<tr>
<td></td>
<td>800</td>
<td>939.2</td>
</tr>
<tr>
<td></td>
<td>1000</td>
<td>945.1</td>
</tr>
<tr>
<td>41.68</td>
<td>107.02</td>
<td></td>
</tr>
<tr>
<td></td>
<td>Saturation*</td>
<td>770.9</td>
</tr>
<tr>
<td></td>
<td>500</td>
<td>784.1</td>
</tr>
<tr>
<td></td>
<td>800</td>
<td>793.1</td>
</tr>
<tr>
<td></td>
<td>1000</td>
<td>798.4</td>
</tr>
<tr>
<td>56.23</td>
<td>133.21</td>
<td></td>
</tr>
<tr>
<td></td>
<td>Saturation*</td>
<td>716.3</td>
</tr>
<tr>
<td></td>
<td>500</td>
<td>729.8</td>
</tr>
<tr>
<td></td>
<td>800</td>
<td>739.2</td>
</tr>
<tr>
<td></td>
<td>1000</td>
<td>745.5</td>
</tr>
<tr>
<td>74.90</td>
<td>166.82</td>
<td></td>
</tr>
<tr>
<td></td>
<td>500</td>
<td>659.1</td>
</tr>
</tbody>
</table>

* Previously established in Ref. 2
Sonic Velocity in Nitrogen Tetroxide. The velocity of sound was measured in saturated liquid N₂O₄ (propellant-grade) over a temperature range of -5 to 56 °C (23 to 132 °F). The resulting values (Table 3 and Fig. 4) were curve fit using a least-squares computer program. The resulting equations, in both metric and English units, are as follows:

\[ c(\text{m/sec}) = 2296 - 4.425 T(K) \quad (\sigma = \pm 3 \text{ m/sec}) \]

and

\[ c(\text{ft/sec}) = 3825 - 8.065 t(\text{F}) \quad (\sigma = \pm 10 \text{ ft/sec}) \]

Adiabatic Compressibility of Nitrogen Tetroxide. The N₂O₄ sonic velocity data were used with saturated liquid N₂O₄ density data (noted in Phase III) to calculate the adiabatic compressibility \( \beta_s \) of N₂O₄. The equations resulting from curve fits of the calculated data are as follows:

\[ \beta_s(\text{atm}^{-1}) = 5.776 \times 10^{-5} + 5.454 \times 10^{-7} t(\text{C}) - 5.185 \times 10^{-9} t(\text{C})^2 \]
\[ -1.659 \times 10^{-11} t(\text{C})^3 + 5.291 \times 10^{-13} t(\text{C})^4 \]

and

\[ \beta_s(\text{psi}^{-1}) = 3.394 \times 10^{-6} + 1.260 \times 10^{-8} t(\text{F}) - 1.486 \times 10^{-10} t(\text{F})^2 \]
\[ -6.329 \times 10^{-13} t(\text{F})^3 + 3.431 \times 10^{-15} t(\text{F})^4 \]

The standard deviations of these curve fits are \( \pm 5.0 \times 10^{-8} \text{ atm}^{-1} \) and \( \pm 3.6 \times 10^{-9} \text{ psi}^{-1} \), respectively. An illustration of a plot of the expression in metric units is shown in Fig. 5.

Sonic Velocity in MHF-3. Measurements of the velocity of sound in MHF-3 (nominal composition of 86 w/o CH₃N₂H₃ - 14 w/o N₂H₄) are being...
TABLE 3

VELOCITY OF SOUND IN SATURATED LIQUID NITROGEN TETROXIDE

<table>
<thead>
<tr>
<th>Temperature</th>
<th>Sonic Velocity</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>C</td>
</tr>
<tr>
<td>-5.33</td>
<td>22.41</td>
</tr>
<tr>
<td>-0.05</td>
<td>31.82</td>
</tr>
<tr>
<td>6.73</td>
<td>44.11</td>
</tr>
<tr>
<td>12.23</td>
<td>54.01</td>
</tr>
<tr>
<td>23.58</td>
<td>74.44</td>
</tr>
<tr>
<td>25.48</td>
<td>77.86</td>
</tr>
<tr>
<td>36.58</td>
<td>97.84</td>
</tr>
<tr>
<td>40.88</td>
<td>105.58</td>
</tr>
<tr>
<td>48.10</td>
<td>118.58</td>
</tr>
<tr>
<td>56.10</td>
<td>132.98</td>
</tr>
</tbody>
</table>
Figure 4. Velocity of Sound in Saturated Liquid Nitrogen Tetroxide

Curve fit from -5 to 56°C with equation:

\[ c_{(m/sec)} = 2296 - 4.425T(K) \]
Figure 5. Adiabatic Compressibility of Saturated Liquid Nitrogen Tetroxide
conducted in the apparatus illustrated in Fig. 2. However, initial testing indicated that the epoxy coating which covered the ultrasonic transducer crystal (Automation Industries No. 57A 2946) used in the existing system was incompatible with MHF-3. A new transducer with a Kel-F protective cover for the ceramic crystal has been ordered.

(C) During the interim, a preliminary sonic velocity of 1604 m/sec (5262 ft/sec) ±4 percent was measured in a MHF-3 sample at 25 C (77 F). This measurement was accomplished by protecting the incompatible epoxy coating with Halocarbon Wax No. 10. At the end of this run, the normally colorless MHF-3 sample had a brownish color which indicated some degree of decomposition and/or reaction. Thus, additional measurements in MHF-3 will be temporarily discontinued until receipt of the new transducer.

Inert-Gas Solubility Measurements

(U) The apparatus for measuring inert-gas solubility in liquid propellants was previously described in detail in Ref. 2. Through the use of this apparatus and the attendant operating procedure, the differential solubility (i.e., the mass of gas dissolved per unit mass of propellant per unit pressure increase) of the gas in the liquid is measured at a particular temperature and pressure. From these data, the total solubility at a particular total pressure can be obtained by the integration of this differential solubility (with respect to pressure) between the propellant vapor pressure (lower limit) and the total pressure (upper limit).

(U) The experimental procedure generally involves the introduction of a known quantity of inert gas into a partially filled propellant cell of known volume. The resulting pressure rise is measured by means of a differential-pressure transducer both before and after the process of solution. These two data serve to establish the distribution of input gas between the ullage and the propellant.
In the original experimental method, the two processes of equilibration (with and without solution) were carried out in succession after a gas input. Since the pressure increments were small, this procedure provided a close approximation to a differential process. However, difficulties in achieving vapor equilibrium within the ullage led to a modification in procedure. The modification (as described in Ref. 3) involved the observation of the two pressure equilibria separately over a sequence of gas inputs. The differential solubilities thus derived are averages over a significant pressure range. This modification has resulted in data of apparently greater reliability, but at a considerable cost in experimental and interpretative effort.

Nitrogen Gas Solubility in Chlorine Pentafluoride. Previous efforts in this program (Ref. 3) had resulted in the measurements of nitrogen gas solubility in ClF$_5$ at three temperatures and pressures. During this quarter, these measurements were extended to a higher temperature and different pressures. The results of all of these measurements are summarized in Table 4. (It should be noted that some of the data previously reported in Ref. 3 have been revised and corrected as a result of the additional measurements.) The estimated accuracy of these data is ±5 percent; a major source of error in this approach arises from taking a small difference between two relatively large, experimentally measured quantities.

These differential solubility data are plotted in Fig. 6 as a function of pressure increase (i.e., the difference between total pressure and the ClF$_5$ vapor pressure at the temperature noted). This difference is the range of integration used in determination of a total solubility. An average value of total pressure for the range shown in Table 4 was assigned to each point.

A linear curve fit of the data shown in Fig. 6 resulted in the following equations:

$$(90 \text{ F}) \Delta S(1 \text{ lb N}_2/1 \text{ lb ClF}_5-\text{psi}) = 1.62 \times 10^{-5} + 0.71 \times 10^{-8} \left[\frac{p(\text{psi})^{-72}}{}\right]$$
TABLE 4

NITROGEN GAS SOLUBILITY IN CHLORINE PENTAFLUORIDE

<table>
<thead>
<tr>
<th>Temperature, F</th>
<th>Vapor Pressure, psia</th>
<th>Total Pressure Range, psia</th>
<th>Differential Solubility, lb N₂/lb CF₅-psi</th>
</tr>
</thead>
<tbody>
<tr>
<td>90</td>
<td>72</td>
<td>430 to 470*</td>
<td>1.89 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>600 to 660</td>
<td>2.02 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>880 to 920</td>
<td>2.21 x 10⁻⁵</td>
</tr>
<tr>
<td>120</td>
<td>115</td>
<td>580 to 610*</td>
<td>2.29 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>780 to 840</td>
<td>2.43 x 10⁻⁵</td>
</tr>
<tr>
<td>150</td>
<td>176</td>
<td>460 to 510</td>
<td>2.50 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>600 to 650</td>
<td>2.56 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>740 to 770*</td>
<td>2.67 x 10⁻⁵</td>
</tr>
<tr>
<td>180</td>
<td>257</td>
<td>380 to 450</td>
<td>2.83 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>670 to 720</td>
<td>2.71 x 10⁻⁵</td>
</tr>
<tr>
<td></td>
<td></td>
<td>840 to 880</td>
<td>2.85 x 10⁻⁵</td>
</tr>
</tbody>
</table>

*Data at these conditions were previously reported in Ref. 3; revisions and corrections to these data have been made on the basis of the additional determinations.
Figure 6. Differential Solubility of Gaseous Nitrogen in Chlorine Pentafluoride

PRESSURE INCREASE, PSI
(TOTAL PRESSURE MINUS CIF$_5$ VAPOR PRESSURE)
These equations have been written in terms of the pressure increase (i.e., total pressure minus ClF$_5$ vapor pressure). The linear curve fits for the lower three temperatures show significant slopes in close agreement. However, at 180°F there is a negligible variation in differential solubility with pressure (according to the least-square criterion). This abrupt change at 180°F could be due to special experimental errors in the higher temperature data, although a re-examination of the experimental results does not support this suspicion.

A total solubility (mass ratio of gas to propellant) as a function of total pressure is obtained by integrating the differential solubility equations over the pressure range between the ClF$_5$ vapor pressure and the total pressure. These integrations result in the following equations:

\[(90°F)\ S(\text{lb} \ N_2/\text{lb} \ ClF_5) = 1.62 \times 10^{-5} + 0.36 \times 10^{-8} \frac{P(\text{psia})^{-72}}{P(\text{psia})^{-72}^2}\]

\[(120°F)\ S(\text{lb} \ N_2/\text{lb} \ ClF_5) = 1.97 \times 10^{-5} + 0.33 \times 10^{-8} \frac{P(\text{psia})^{-115}}{P(\text{psia})^{-115}^2}\]

\[(150°F)\ S(\text{lb} \ N_2/\text{lb} \ ClF_5) = 2.30 \times 10^{-5} + 0.31 \times 10^{-8} \frac{P(\text{psia})^{-176}}{P(\text{psia})^{-176}^2}\]

\[(180°F)\ S(\text{lb} \ N_2/\text{lb} \ ClF_5) = 2.80 \times 10^{-5} \frac{P(\text{psia})^{-257}}{P(\text{psia})^{-257}}\]

The total solubilities given by these equations have been graphically illustrated in Fig. 7.
Figure 7. Total Solubility of Gaseous Nitrogen in Chlorine Pentafluoride

PRESSURE INCREASE, PSI  
(TOTAL PRESSURE MINUS CIF₅ VAPOR PRESSURE)
Specific Heat Measurements

(U) During the current report period, efforts under the specific heat measurement task were directed at: (1) completion of specific heat measurements on the MHF-3 fuel blend, and (2) extension of the lower temperature limit of the MHF-5 fuel blend specific heat data. These experimental determinations were conducted in an adiabatic calorimeter, which has been previously described in Ref. 1, 3 and 6.

(C) Specific Heat of MHF-3

Specific heat measurements on the MHF-3 fuel blend (nominal composition of 86 w/o CH$_3$N$_2$H$_3$-14 w/o N$_2$H$_4$) were successfully completed on the liquid phase over the temperature range of -46.0 to 47.3 C (-50.8 to 117.2 F) using a calibrated sample container described in Ref. 3. The experimental specific heat data, which are listed in Table 5, are somewhat higher (e.g., ~2 percent greater at 20 C) than those data calculated from additive contributions of individual component specific heats. Through the use of a least-squares computer program, the data were curve fit over the given temperature range with the following equations:

\[ C_p(\text{cal/gm-C}) = 0.708 + 3.807 \times 10^{-4} t(C) + 2.110 \times 10^{-6} t(C)^2 \]

and

\[ C_p(\text{Btu/lb-F}) = 0.702 + 1.698 \times 10^{-4} t(F) + 6.512 \times 10^{-7} t(F)^2 \]

Figure 8 illustrates a graphical representation of the curve fit in the metric units. The average scatter of the experimental data from this smoothed specific heat-temperature curve is ±0.4 percent.

(U) Results of chemical analysis of the MHF-3 sample used in the specific heat measurements are also shown in Table 5.
# TABLE 5

EXPERIMENTAL SPECIFIC HEAT OF MHF-3 FUEL BLEND*

<table>
<thead>
<tr>
<th>Temperature</th>
<th>Specific Heat (cal/gm·C)</th>
</tr>
</thead>
<tbody>
<tr>
<td>-46.01</td>
<td>0.695</td>
</tr>
<tr>
<td>-44.48</td>
<td>0.693</td>
</tr>
<tr>
<td>-43.96</td>
<td>0.691</td>
</tr>
<tr>
<td>-39.90</td>
<td>0.692</td>
</tr>
<tr>
<td>-38.33</td>
<td>0.698</td>
</tr>
<tr>
<td>-37.86</td>
<td>0.704</td>
</tr>
<tr>
<td>-36.32</td>
<td>0.705</td>
</tr>
<tr>
<td>-29.74</td>
<td>0.700</td>
</tr>
<tr>
<td>-25.37</td>
<td>0.700</td>
</tr>
<tr>
<td>-17.82</td>
<td>0.699</td>
</tr>
<tr>
<td>-15.83</td>
<td>0.698</td>
</tr>
<tr>
<td>-8.92</td>
<td>0.704</td>
</tr>
<tr>
<td>-6.95</td>
<td>0.709</td>
</tr>
<tr>
<td>0.32</td>
<td>0.705</td>
</tr>
<tr>
<td>1.85</td>
<td>0.709</td>
</tr>
<tr>
<td>2.30</td>
<td>0.712</td>
</tr>
<tr>
<td>3.83</td>
<td>0.707</td>
</tr>
<tr>
<td>11.46</td>
<td>0.719</td>
</tr>
<tr>
<td>13.41</td>
<td>0.715</td>
</tr>
<tr>
<td>18.98</td>
<td>0.714</td>
</tr>
<tr>
<td>20.96</td>
<td>0.721</td>
</tr>
<tr>
<td>25.43</td>
<td>0.715</td>
</tr>
<tr>
<td>27.36</td>
<td>0.721</td>
</tr>
<tr>
<td>32.64</td>
<td>0.723</td>
</tr>
<tr>
<td>34.58</td>
<td>0.724</td>
</tr>
<tr>
<td>40.60</td>
<td>0.724</td>
</tr>
<tr>
<td>42.52</td>
<td>0.737</td>
</tr>
<tr>
<td>44.76</td>
<td>0.730</td>
</tr>
<tr>
<td>47.31</td>
<td>0.727</td>
</tr>
</tbody>
</table>

*Sample Composition:

- CH$_3$N$_2$H$_7$, w/o: 84.7
- N$_2$H$_4$, w/o: 14.0
- H$_2$O, w/o: 1.0
- NH$_3$, w/o: 0.1
- CH$_3$NH$_2$: trace
- Other soluble impurities, w/o: <0.2
Prior to conducting MHF-3 specific heat measurements above room temperature, thermal stability tests were initiated to ensure that vaporization of the liquid sample, and/or its potential decomposition products (as indicated by a pressure buildup) at the higher temperatures would not introduce significant errors in the measurements. A gas evolution apparatus was fabricated. The apparatus consisted of a copper sample container joined by means of a Kovar-to-Pyrex seal to a Hg-filled glass manometer in line with another manometer opened to ambient pressure. Using an ullage comparable to that maintained in the specific heat sample container at 75°C (167°F), gas evolution tests were conducted with MHF-3 at that temperature. The resulting average pressure buildup of ~0.3 mm Hg/hr indicated a rate which was tolerable during the period required for the specific heat measurements.

Measurements of specific heat of the MHF-5 fuel blend (nominal composition of 55 w/o CH₃N₂, 26 w/o N₂H₄, 19 w/o N₂H₄·HNO₃) were successfully completed on the liquid phase over the temperature range of -47.4 to 7.5°C (-53.3 to 45.5°F). The sample container, which was fabricated and calibrated for use with MHF-5 (Ref. 3), was also used in the MHF-5 measurements. Because of the potential thermal sensitivity of MHF-5, the silver solder brazing normally used to cap (to ensure a complete seal) the sample container filling tube was replaced with indium metal, which has a lower melting point (~310°F). The sample container was loaded with 64.02 grams of MHF-5 under dry nitrogen in a glove bag.

The experimental specific heat data and the results of chemical analysis of the MHF-5 sample are listed in Table 6. These data have been correlated with higher-temperature data (Ref. 7) and curve fit over a temperature range of -47.4 to 68.0°C (-53.3 to 154.4°F). The resulting equations:

\[ C_p(\text{cal/gm-C}) = 0.664 + 1.21 \times 10^{-4} t(C) \]

and

\[ C_p(\text{Btu/1b-F}) = 0.662 + 6.74 \times 10^{-5} t(F) \]
### EXPERIMENTAL SPECIFIC HEAT OF NH-FUEL BLEND

<table>
<thead>
<tr>
<th>Temperature C</th>
<th>Temperature F</th>
<th>Specific Heat cal/gm-C</th>
</tr>
</thead>
<tbody>
<tr>
<td>-47.38</td>
<td>-53.28</td>
<td>0.666</td>
</tr>
<tr>
<td>-45.24</td>
<td>-49.43</td>
<td>0.657</td>
</tr>
<tr>
<td>-38.64</td>
<td>-37.55</td>
<td>0.656</td>
</tr>
<tr>
<td>-36.62</td>
<td>-33.92</td>
<td>0.662</td>
</tr>
<tr>
<td>-36.52</td>
<td>-33.74</td>
<td>0.663</td>
</tr>
<tr>
<td>-29.84</td>
<td>-21.71</td>
<td>0.661</td>
</tr>
<tr>
<td>-27.72</td>
<td>-17.90</td>
<td>0.659</td>
</tr>
<tr>
<td>-20.66</td>
<td>-5.19</td>
<td>0.665</td>
</tr>
<tr>
<td>-13.31</td>
<td>8.04</td>
<td>0.667</td>
</tr>
<tr>
<td>-11.21</td>
<td>11.82</td>
<td>0.659</td>
</tr>
<tr>
<td>-4.50</td>
<td>23.90</td>
<td>0.663</td>
</tr>
<tr>
<td>-2.38</td>
<td>27.72</td>
<td>0.667</td>
</tr>
<tr>
<td>5.39</td>
<td>41.70</td>
<td>0.662</td>
</tr>
<tr>
<td>7.51</td>
<td>45.52</td>
<td>0.671</td>
</tr>
</tbody>
</table>

*Sample Composition:

CH$_3$N$_2$H$_3$, w/o 54.9
N$_2$H$_4$, w/o 25.4
N$_2$H$_4$:HNO$_3$, w/o 18.9
NH$_3$, w/o 0.2
H$_2$O, w/o 0.6
indicating a standard deviation of 0.005 cal/gm-C (Btu/lb-F). A graphical representation of the data in metric units is presented in Fig. 9.

Although the data obtained at Rocketdyne resulted in a curve fit value at 7°C (44.6°F) which is 1.3 percent greater than a corresponding value from a previously reported (Ref. 8) curve fit, the analytical correlation of both sets of actual experimental data indicates good agreement of all of the data with the above equations.

Prior to specific heat measurements on MHF-5, gas evolution tests were conducted to characterize any potential pressure buildup problems. The gas evolution apparatus used previously with MHF-3 was reassembled, leak checked, and loaded with MHF-5 for tests at ambient temperatures. Using an ullage of half that maintained in the specific heat sample container at ambient temperatures, the gas evolution test resulted in an average pressure buildup of ~1.8 mm Hg/hr. For the period of time involved in calorimetric measurements, this rate would not produce significant errors in the determinations.

Thermal Conductivity Measurements

Thermal conductivity measurements of liquid propellants have been conducted through the use of a steady-state, concentric-cylinder conductivity cell described previously in Ref. 1 and 2. With this apparatus, the test liquid is contained in a very thin, annular passage between two aluminum alloy cylinders. An electrical resistance heater, located at the center of the inner cylinder, supplies the heat energy to establish a temperature gradient across the liquid layer. Copper constantan thermocouples, embedded in the cylinders close to the surfaces of the test fluid cavity, are used to measure the temperature gradient. A regulated d-c power supply is employed for the cell heater power.

In evaluation of the applicability of this apparatus to measurements involving selected oxidizers, preliminary analysis indicates that replacement
Figure 9. Specific Heat of MHF-5

CURVE FIT FROM -47 TO 68°C WITH EQUATION:

\[ c_p \text{(cal/gm-c)} = 0.664 + 1.21 \times 10^{-4} \ t(c) \]
and/or redesign of the cell sealing surfaces may be required. In addition, other techniques (notably those involving transient measurement procedures) are being evaluated under in-house funding to provide an alternate technique if the present apparatus is not suitable. If the present apparatus is considered applicable to oxidizer measurements, the apparatus will be completely disassembled and cleaned to remove all traces of residual fuel.

(U) The selection of ClF₅ as the initial oxidizer of interest for experimental thermal conductivity characterization has been approved by the Air Force Project Officer.

Viscosity Measurements

(U) Viscosity measurements on liquid propellants are being conducted with a stainless-steel capillary viscometer which has been described in Ref. 2. Kinematic viscosity is obtained by timing the decay of a small liquid head in flow through the capillary tubing. The movement of a liquid is followed by means of a differential transformer which determines the position of a magnetic float (indicating the liquid level) within a reservoir in the viscometer. For streamline flow in the capillary, the apparatus requires a single calibration constant which is obtained from tests with distilled water at various temperature levels.

(U) Viscosity of Chlorine Pentafluoride. Previously in this program (Ref. 3), ClF₅ viscosity measurements had been conducted at five temperature levels from 75 to 176°F at or near the corresponding ClF₅ vapor pressure. During this quarter, viscosity measurements on ClF₅ were continued to provide high-temperature data and investigate the possible effect of pressurization on viscosity. The results from these measurements, which were conducted at 202 ±1°F (94.4 ±0.6°C) over a total pressure range of 360 to 910 psia, are shown in Table 7. These total pressures were obtained through the use of gaseous nitrogen in the system ullage. The pressurizing gas was introduced shortly before each run to avoid solution equilibrium with the propellant.
TABLE 7

EXPERIMENTAL CHLORINE PENTAFLUORIDE VISCOSITY DATA*

<table>
<thead>
<tr>
<th>Total Pressure, psia</th>
<th>Kinematic Viscosity, centistokes</th>
</tr>
</thead>
<tbody>
<tr>
<td>360</td>
<td>0.114</td>
</tr>
<tr>
<td>380</td>
<td>0.116</td>
</tr>
<tr>
<td>570</td>
<td>0.114</td>
</tr>
<tr>
<td>720</td>
<td>0.114</td>
</tr>
<tr>
<td>770</td>
<td>0.112</td>
</tr>
<tr>
<td>810</td>
<td>0.114</td>
</tr>
<tr>
<td>860</td>
<td>0.110</td>
</tr>
<tr>
<td>910</td>
<td>0.118</td>
</tr>
</tbody>
</table>

*Measurements conducted at 202 ±1 F; ClF$_5$ vapor pressure is 332 psia.

(U) These results indicate that in the range of 360 to 910 psia, pressure has no effect on the kinematic viscosity of ClF$_5$. Deviations shown in these measurements (notably at the two highest pressures) are considered to be spurious experimental effects. The source of these deviations, and their possible relation to high-pressure operation, is not presently known. In previous measurements on the saturated liquid (at temperatures up to 176 F), the viscometer has performed with a precision on the order of ±1 percent.

(U) The kinematic viscosity of ClF$_5$ at 202 F is 0.114 centistokes for the saturated and compressed liquid below 1000 psia. For the saturated liquid, using a ClF$_5$ density established in Ref. 6, the corresponding absolute or dynamic viscosity is 0.168 centipoises. This value has been incorporated in Fig. 10, which illustrates previously available ClF$_5$ viscosity data.
Figure 10. Viscosity of Saturated Liquid Chlorine Pentafluoride
The invariance of kinematic viscosity with relatively minor compression of the liquid is in accordance with general theoretical and empirical correlations. The absolute viscosity of the compressed liquid must be somewhat larger than that at saturation and it is a function of the isothermal compressibility of the liquid (which is not known for CIF$_5$ at 200°F). Even at this high temperature (relative to the critical point) and correspondingly large compressibility, the pressure effects are not significant unless very high pressures are attained. From the data in Ref. 2, the adiabatic compressibility of CIF$_5$ at 202°F is estimated at about $4 \times 10^{-5}$ psi$^{-1}$ (with the isothermal compressibility being typically larger by 10 to 50 percent). Hence, the density increase and corresponding absolute viscosity increase within 1000 psi should be in the order of 3 percent.
PHASE III: EVALUATION AND COMPILATION OF DATA

OBJECTIVE

(U) During the entire period of the contractual program, efforts under Phase III are directed toward the assembly of all data generated by Phases I and II, verification of all the data sources, critical comparison of conflicting data, and tabulation of the results. The final selection of the best values will be summarized in an annual technical report.

RESULTS AND ACCOMPLISHMENTS

(U) In the current report period, Phase III effort has been directed primarily at the compilation and evaluation of physical properties of various selected propellants from the references located in the Phase I literature survey. The remaining effort has been concentrated on the reduction, evaluation, and correlation of all data generated as a result of the Phase II experimental efforts; however, to maintain continuity in this text, these results are reported under pertinent Phase II headings.

Surface Tension of Propellants

(U) At the request of the Air Force Project Engineer, all previously published surface tension data on \( \text{ClF}_3 \), \( \text{ClF}_5 \), \( \text{N}_2\text{O}_4 \), \( 50\text{N}_2\text{H}_4-50\text{UDMH} \), \( \text{N}_2\text{H}_4 \), \( \text{MMH} \), \( \text{UDMH} \), \( \text{MHF}-3 \), and \( \text{MHF}-5 \) have been compiled and evaluated. These data are summarized in the following paragraphs.

(U) Chlorine Trifluoride. The surface tension of saturated liquid \( \text{ClF}_3 \) has been experimentally determined (Ref. 9) by the method of capillary rise over the temperature range of 0 to 50 °C (32 to 122 °F). A curve fit of the data, shown in Fig. 11, can be expressed by the following equations:

\[
\gamma(\text{dynes/cm}) = 26.7 - 0.16 t(\circ) 
\]
and

\[ \gamma(Lb/ft) = 2.92 \times 10^{-3} - 6.1 \times 10^{-6} T(F) \]

(U) Chlorine Pentafluoride. The surface tension of \( ClF_5 \), which was experimentally determined (Ref. 6) over a temperature range of -50 to 25 C (-58 to 77 F), is shown in Fig. 12. The equations representing these data are:

\[ \gamma(dynes/cm) = 57.949 - 0.1446 T(K) \]

and

\[ \gamma(Lb/ft) = 3.9708 \times 10^{-3} - 0.5506 \times 10^{-5} T(F) \]

(U) Nitrogen Tetroxide. The surface tension of liquid \( N_2O_4 \) was experimentally determined to be 27.0 dynes/cm \( (1.850 \times 10^{-3} \text{ lb/ft}) \) at 20 C \( (68 \text{ F}) \) with a duNouy tensiometer (Ref. 10). This comparatively recent data point conflicts with previously reported data (Ref. 11); however, this value represents the propellant-grade \( N_2O_4 \) presently in use.

(U) 50 w/o \( N_2H_4 \)-50 w/o UDMH. A duNouy tensiometer was also used to measure (Ref. 10) the surface tension of the 50 w/o \( N_2H_4 \)-50 w/o UDMH fuel blend at 20C \( (68 \text{ F}) \). A value of 30.3 dynes/cm \( (2.076 \times 10^{-3} \text{ lb/ft}) \) was reported from this measurement.

(U) Hydrazine. Although surface tension values of 66.67 dynes/cm at 25 C and 62.32 dynes/cm at 35 C had been previously reported for anhydrous \( N_2H_4 \) in Ref. 12 and 13, respectively, the later work of Ref. 14 was used to curve fit the data shown in Fig. 13. The equations obtained from the curve fit are as follows:

\[ \gamma(dynes/cm) = 72.46 - 0.242 T(C) \]
Figure 11. Surface Tension of Saturated Liquid Chlorine Trifluoride

CURVE FIT FROM 0 TO 50°C WITH EQUATION
\[ \gamma (\text{dynes/cm}) = 26.7 - 0.8 \times 10^{-2} \times \text{C} \]
Figure 12. Surface Tension of Saturated Liquid Chlorine Pentafluoride (Ref. 6)
Figure 13. Surface Tension of Anhydrous Hydrazine (Ref. 14).
and
\[ \gamma_{(1b/ft)} = 5.261 \times 10^{-3} - 9.25 \times 10^{-6} t_F \]

(U) **Monomethylhydrazine.** The surface tension of \( \text{CH}_3\text{N}_2\text{H}_3 \) (MMH) was measured by the capillary rise method over the temperature range of -40 to 60 C (-40 to 140 F) in Ref. 15. These data (Fig. 14) were curve fit with the following equations:

\[ \gamma_{(\text{dynes/cm})} = 36.29 - 0.0987 t_C \]

and
\[ \gamma_{(\text{lb/ft})} = 2.607 \times 10^{-3} - 3.76 \times 10^{-6} t_F \]

(U) **Unsymmetrical Dimethylhydrazine.** The surface tension of \( (\text{CH}_3)_2\text{N}_2\text{H}_2 \) (UDMH) was also measured by the capillary rise method over the temperature range of -54 to 50 C (-65.2 to 122 F) in Ref. 15. These data (Fig. 15) can be represented with the following equations:

\[ \gamma_{(\text{dynes/cm})} = 26.53 - 0.108 t_C + 2.08 \times 10^{-4} t_C^2 \]

and
\[ \gamma_{(\text{lb/ft})} = 1.954 \times 10^{-3} - 4.399 \times 10^{-6} t_F + 4.390 \times 10^{-9} t_F^2 \]

A value of 24.4 dynes/cm (1.612 \( \times 10^{-3} \) lb/ft) at 20 C (68 F), reported in Ref. 10, is in good agreement with these data.

(C) **MHF-3 Fuel Blend.** No data have been reported on the surface tension of the MHF-3 fuel blend (nominal composition of 86 w/o MMH-14 w/o \( \text{N}_2\text{H}_4 \)).

(C) **MHF-5 Fuel Blend.** No experimental surface tension data have been reported for the MHF-5 fuel blend (nominal composition of 55 w/o MMH-26 w/o \( \text{N}_2\text{H}_4 \)-19 w/o \( \text{N}_2\text{H}_5\text{N}_0_3 \)).
Figure 13. Surface Tension of Anhydrous Hydrazine (Ref. 14).
and
\[ \gamma_{\text{dyne/cm}} = 36.29 - 0.0987 t(\degree C) \]

and
\[ \gamma_{\text{lb/ft}} = 2.607 \times 10^{-3} - 3.76 \times 10^{-6} t(\degree F) \]

(U) Monomethylhydrazine. The surface tension of \( \text{CH}_3\text{N}_2\text{H}_3 \) (MMH) was measured by the capillary rise method over the temperature range of -40 to 60 \( \degree C \) (-40 to 140 \( \degree F \)) in Ref. 15. These data (Fig. 14) were curve fit with the following equations:

\[ \gamma_{\text{dyne/cm}} = 26.53 - 0.108 t(\degree C) + 2.08 \times 10^{-4} t(\degree C)^2 \]

and
\[ \gamma_{\text{lb/ft}} = 1.954 \times 10^{-3} - 4.399 \times 10^{-6} t(\degree F) + 4.390 \times 10^{-9} t(\degree F)^2 \]

A value of 24.4 dynes/cm (1.612 \( \times 10^{-3} \) lb/ft) at 20 \( \degree C \) (68 \( \degree F \)), reported in Ref. 10, is in good agreement with these data.

(C) MHF-3 Fuel Blend. No data have been reported on the surface tension of the MHF-3 fuel blend (nominal composition of 86 w/o MMH-14 w/o \( \text{N}_2\text{H}_4 \)).

(C) MHF-5 Fuel Blend. No experimental surface tension data have been reported for the MHF-5 fuel blend (nominal composition of 55 w/o MMH-26 w/o \( \text{N}_2\text{H}_4 \)-20 w/o \( \text{N}_2\text{H}_5\text{NO}_3 \)).
Figure 14. Surface Tension of Monomethyldiazine (Ref. 15)
33.0
32.0
31.0
30.0
29.0
28.0
27.0
26.0
25.0
24.0
23.0
22.0
21.0
-60 -50 -40 -30 -20 -10 0 10 20 30 40 50
TEMPERATURE, C

SURFACE TENSION, DYNES/CM

Figure 15. Surface Tension of UDMH (Ref. 15)

CURVE FITTED FROM -54 TO 50 C WITH EQUATION:

\[ \gamma (\text{DYNES/CM}) = 26.53 - 0.108t(C) + 2.08 \times 10^{-4} t(C)^2 \]
Nitrogen Tetroxide Physical Property Compilations

Preparation of a physical properties bibliography for N₂O₄ is continuing. In addition, a complete evaluation of N₂O₄ physical properties data is in progress.

Nitrogen Tetroxide Density. As a part of the N₂O₄ physical property compilations, the available experimental data on N₂O₄ density was compiled and correlated. These correlated data were used in the calculation of the adiabatic compressibility of liquid N₂O₄ from experimental sonic velocity data (as noted in Phase II). From a literature survey, saturated liquid N₂O₄ density data were obtained over the range -11.2 to 158.2 C (11.8 to 316.8 F). These data (Ref. 16 through 20) were correlated and curve fit with a least-squares computer program from -11.2 to 93.3 C (11.8 to 199.9 F). The following equations were obtained from this curve fit:

\[
\rho_{(\text{g/m/cc})} = 1.490 - 2.124 \times 10^{-3} t(\text{C}) - 5.01 \times 10^{-6} t(\text{C})^2
\]

and

\[
\rho_{(\text{lb/cu ft})} = 95.31 - 6.749 \times 10^{-2} t(\text{F}) - 9.66 \times 10^{-5} t(\text{F})^2
\]

The available data above 93.3 C were omitted from the present correlation to simplify the curve fit; however, this temperature range satisfied the range of the available sonic velocity data.
REFERENCES


2. AFRPL-TR-67-151, Annual Summary Report, Engineering Property Data on Rocket Propellants, Contract AF04(611)-11407, Rocketdyne, a Division of North American Aviation, Inc., Canoga Park, California, June 1967. CONFIDENTIAL.


7. Private communication, C. J. Grelecki, RMD, to K. J. Youel, Rocketdyne, 2 October 1967 (Experimental data points from work reported in Ref. 8).

8. RMD 5046-F, Advanced Propellants Investigation for Prepackaged Liquid Engines, Contract No. N600(19) 62259, Reaction Motors Division, Thiokol Corporation, Denville, New Jersey, June 1965, CONFIDENTIAL.


The results of the fifth quarter year's effort of a 24-month program on the analytical and experimental characterization of the physical properties of selected liquid propellants are presented in three phases. In Phase I, a continuous review of the literature was conducted to ensure the acquisition and documentation of the latest possible propellant properties data for evaluation and possible inclusion into a propellant properties handbook. Phase II experimental efforts have resulted in the measurement of chlorine trifluoride vapor pressure in the high-temperature region and an estimation of critical properties; sonic velocity in chlorine trifluoride and chlorine pentafluoride under pressurized conditions; nitrogen gas solubility in chlorine pentafluoride; specific heat of saturated liquid chlorine trifluoride; thermal conductivity of MHF-5; and viscosity of chlorine pentafluoride. Phase III efforts included the evaluation and assembly of all data generated in Phases I and II, a curve fit of specific heat data on solid chlorine trifluoride, and a report of unpublished data on the normal boiling point of chlorine pentafluoride and chlorine pentafluoride storability. (U)
<table>
<thead>
<tr>
<th>KEY WORDS</th>
<th>LINK A</th>
<th>LINK B</th>
<th>LINK C</th>
</tr>
</thead>
<tbody>
<tr>
<td>Propellants</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Chlorine Trifluoride</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Chlorine Pentafluoride</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Hydrazine</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Sonic Velocity</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Thermal Conductivity</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Specific Heat</td>
<td></td>
<td></td>
<td></td>
</tr>
</tbody>
</table>
THIS REPORT HAS BEEN DELIMITED AND CLEARED FOR PUBLIC RELEASE UNDER DOD DIRECTIVE 5200.20 AND NO RESTRICTIONS ARE IMPOSED UPON ITS USE AND DISCLOSURE.

DISTRIBUTION STATEMENT A

APPROVED FOR PUBLIC RELEASE;
DISTRIBUTION UNLIMITED.