Improved Viscous Brake Design Using Simulation

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The SDIO High Altitude Balloon Experiment (HABE) required a rapid controlled descent of a large, heavy payload from a balloon platform. An existing viscous brake concept was chosen for the job, but the short time and space available coupled to the heavy payload resulted in very high power densities. Existing rules of thumb for viscous brakes were insufficient for the design. Viscous brake phenomenology and the particular physical constraints were coded into a modern continuous simulation applications package. After investigating both full dynamic and quasi-steady models, it was decided that the latter was adequate for all but the first and last small fraction of the descent. The model includes temperature dependent viscosity and a heat transfer model. The model been used extensively to investigate the design space and to develop test plans.

Simulation, Viscous Brake, Thermal Mechanical Model, Viscous Brake Design
IMPROVED VISCOS BRAKE DESIGN USING SIMULATION

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ABSTRACT
The SDIO High Altitude Balloon Experiment (HABE) required a rapid controlled descent of a large, heavy payload from a balloon platform. An existing viscous brake concept was chosen for the job, but the short time and space available coupled to the heavy payload resulted in very high power densities. Existing rules of thumb for viscous brakes were insufficient for the design. Viscous brake phenomenology and the particular physical constraints were coded into a modern continuous simulation applications package. After investigating both full dynamic and quasi-steady models, it was decided that the latter was adequate for all but the first and last small fraction of the descent. The model includes temperature dependent viscosity and a heat transfer model. The model was used extensively to investigate the design space and to develop test plans.

BACKGROUND
A unique ascent procedure has been proposed for the HABE experiment. At liftoff, all of the helium will be contained in a "tow balloon". At a designated altitude, the payload will be dropped approximately 500 feet below the tow balloon, allowing the deployment of the main balloon and parachute which are attached to the tow and stored at the top of the payload section. See Figure 1 for a representation of the reel down sequence.

The design of the system is constrained by the following: full reel down must be accomplished within 3.5 minutes to insure that the main balloon is fully deployed before gas begins to flow from the tow balloon to the main balloon; at the end of the 500ft reel down, separation velocity must be low enough to avoid excessive stress on the main balloon and parachute which must provide the final breaking force; the fluid temperature must not exceed 450°F to avoid thermal breakdown. A desired capability is that the system should stay within these constraints for initial conditions within the range from 0°F to 100°F.

The Aerospace Engineering Division of the Phillips Laboratory has had extensive experience using reel down systems with integrated viscous brakes (Wagner and Doherty 1958). The requirements for the HABE mission are much more severe in payload weight and time available for the deployment. This causes very high brake power loads, resulting in significant heating of the brake fluid which, in turn, substantially lowers fluid viscosity. If the brake fluid viscosity is too low, thermal runaway could result. If it is too high, reel down will take too long. A mathematical simulation of the deployment was required to assist in the design and test of the reel down system.

THEORY
The motion of the system is determined by applying Newton's 2nd Law to each point mass:

\[ m \frac{d^2R}{dt^2} = \sum F_i \]  (1)

Where: \( m \) = mass
\( R \) = position vector to center of mass
\( F_i \) = the \( i \)th force vector

Since reel down would occur at some time after liftoff, it is assumed that the system would be traveling horizontally at the speed of the wind, and that any horizontal relative motion between balloon and payload would be damped out. This reduces the scope of the analysis to the vertical coordinate only.

A free body diagram of each mass is shown in Figure 2. The equation of motion for each mass is:

\[ m_1 \frac{d^2y_1}{dt^2} = B - D - T - (M_1 + \Delta M) \]  (2)

\[ m_2 \frac{d^2y_2}{dt^2} = T - (M_2 + \Delta M) \]  (3)

Where:
\( T \) = Tension in the connection
\( W \) = Weight
\( B \) = Buoyancy
\( D \) = Drag

Subscripts:
1 = Balloon and Helium
2 = Payload

Figure 1. The reel down sequence.
and where \( \Delta W \) the weight of the main balloon and parachute that has been extracted from the payload section. This added weight on the upper mass should include the additional force required to accelerate the main balloon and parachute from the velocity of the payload to the velocity of the upper mass. This tension is equal to the instantaneous rate of change of momentum of those items which is negligible for this problem.

\[
\begin{align*}
T_{\text{brake}} = J \frac{d^2 \theta}{dt^2} + TR - M_{\text{brake}}
\end{align*}
\]

Where: 
- \( J \) = moment of inertia of the rotating system (slug \( \text{ft}^2 \))
- \( \theta \) = angle of rotation (rad)
- \( R \) = radius to cable on reel (ft)
- \( M_{\text{brake}} \) = moment caused by brake (lb \( \text{ft} \))

The brake moment (Wagner and Doherty 1958) is:

\[
M_{\text{brake}} = a \pi \rho \omega (r_i^2 - r_o^2)/d
\]

Where:
- \( v \) = kinetic viscosity (\( \text{ft}^2/\text{s} \))
- \( \rho \) = brake fluid density (slugs/\( \text{ft}^3 \))
- \( \omega \) = angular velocity, same as \( d\theta/dt \) (rad/\( \text{s} \))
- \( a \) = number of rotors in the brake
- \( r_i \) = outer or inner radius of active brake surface (ft)
- \( d \) = distance between disk rotors and stators (ft)

Figure 3. Isometric of brake and reels.

As stated earlier, the HABE brake will be absorbing a lot of power, which will cause fluid heating. The temperature dependence of viscosity for many liquids is known to follow the following relationship:

\[
v = A v_n \exp(B/T)
\]

Where \( v_n \) is the nominal viscosity for the fluid and \( A \) and \( B \) are constants for the material. For the GE VISCASIL fluids used in the HABE brakes, the constants were found to be 0.003 and 1740 respectively (when \( T \) is measured in K).

The reel down is accomplished by spooling DuPont KEVLAR® cable from two straight-sided spools. Assuming a continuous reduction of radius with length, the instantaneous radius is:

\[
R = \sqrt{R_o^2 - (R_i^2 - R_f^2)/L}
\]

Where: 
- \( L \) is the instantaneous length
- \( L \) is the total length
- \( R_{o} \) is the outer radius of active masses
- \( R_{i} \) is the inner radius of active masses
- \( R_{f} \) is the final radius of active masses

and subscript 0 refers to the outer extent of the cable radius when fully wound on the spool, and i is the radius when fully deployed.

The rate of change of length of cable is related to the turning of the cable by the rotational formula:

\[
\frac{dl}{dt} = R \frac{d\theta}{dt}
\]

If the cable is assumed to be have infinite stiffness, then the separation of the two masses is exactly determined by the amount of reel down, which reduces the degrees of freedom of the problem to 2. This seems especially justified since DuPont KEVLAR® is a very stiff material. With infinite stiffness, the length rolled off the spool, \( l \), is directly related to the distance between the two masses by the equation:

\[
l = \frac{1}{2} \sqrt{(R_o^2 - R_i^2)}
\]
We obtain: monotonically decreasing with time as shown in the torque from the brake. After solving for decreasing with time, the torque on the brake is from the reel down cables is exactly balanced by weight and the radius of the line on the reel sum of the moments is zero. That is, the torque deployed in 200 seconds. With both the payload to zero, which yields the static result that the viscosity of 125,000 centistokes, the 500 foot line obtained by setting the second derivative of Equation 4. The steady state solution of Equation 4 is required for acceptable operation of the system.

Steady State Solution: The steady state program was run with various temperature and other variables. This dissipated power appears as heat in the brake fluid. As the fluid heats up, it transfers heat to the rotors and stators which, in turn, pass it on to the housing and the shaft. The heating of the brake was investigated by, first, formulating a detailed thermal model of the brake, then using the results of that model to guide in the formulation of a simpler model for incorporation in the ACSL solution. That model was a three node model which consisted of three equations of the form:

\[ M C_i \frac{dT_i}{dt} = \sum_{j \neq i} K_{ij} (T_j - T_i) + Q_i \]  

Where: \( T_i \) = temperature of the i\textsuperscript{th} node

\( K_{ij} \) = thermal conductance from i\textsuperscript{th} to j\textsuperscript{th} node

\( Q_i \) = internal heat generation for the i\textsuperscript{th} node

\( M_i \) = mass of the i\textsuperscript{th} node

\( C_i \) = heat capacity of the i\textsuperscript{th} node

The nodes represent the following: 1, the fluid, which was the only node to have any internal heat generation, 2, the rotors and stators, and 3 the housing and shaft. The values used for thermal mass (M\textsubscript{C}) were 3,990J/K for 1, 21,100J/K for 2, and 14,600J/K for 3 (Mironer 1992). The conductance between 1 and 2 was 680W/K, between 1 and 3 was 47.5W/K, and between 2 and 3 was 9.50W/K. The internal heat generation is the power dissipation computed in Equation 15.

INITIAL DESIGN CALCULATIONS

The steady state program was run with various nominal fluid viscosities to determine what was required for acceptable operation of the system. With an initial temperature of 60°F, and a nominal viscosity of 125,000 centistokes, the 500 foot line deployed in 200 seconds. With both the payload weight and the radius of the line on the reel decreasing with time, the torque on the brake is monotonically decreasing with time as shown in

\[ \omega = \frac{TR_d}{n \pi \nu \rho \left( r_i^2 - r_f^2 \right)} \]
The best option for calibration of the thermal force input, space vehicle trajectories, and thermal, mechanical requirement and a decreasing, then increasing interests include the simulation of atmospheric and pull, then increased dramatically near the midpoint the United States Military Academy and Worcester operation resulted in high initial force which Technology, Wright Patterson AFB, OH. He taught at proportional to the speed curve. Constant speed Engineering in 1975 from the Air Force Institute of the expected operational performance shown in Figure 9. This particular case is judged successful since brake fluid temperature, total time, and final rollout velocity were within the design constraints. This viscosity will behave differently at different starting temperatures, as shown in Figure 10.

PROTOTYPE TESTING
At this point it is proper to point out that the model described above, while sophisticated for standard brake design practice, contains many assumptions and simplifications which lead to inaccuracies in the final results. Some of the major uncertainties are the small differences in geometry from one rotor-stator working surface to the next due to manufacturing tolerances, the actual viscosity of the fluid at the shear rates of the test, the accuracy of a single average temperature in determining the correct average viscosity of the fluid in the brake, the accuracy of the 3 node model for predicting the heat transfer events of the brake, and the accuracy of the thermal masses and the thermal conductances in the model. A horizontal brake reel roll out test was planned to insure that the system in place and another truck pulled out the Kevlar Line. Since the trucks were available for only a limited amount of time, and since each test required at least 2 hours for cool down of the brake and rewinding of the line on the spools, the probable number of tests is from 2 to 4. In order to design the test for maximum useful data, several possible test runs were simulated with the system model. Accurate speedometer and load cell readout was available in the truck cab.

Two trucks were used to perform the horizontal brake reel roll out test. One truck was in the system in place and another truck pulled out the Kevlar Line. Since the trucks were available for only a limited amount of time, and since each test required at least 2 hours for cool down of the brake and rewinding of the line on the spools, the probable number of tests is from 2 to 4. In order to design the test for maximum useful data, several possible test runs were simulated with the system model. Accurate speedometer and load cell readout was available in the truck cab.

The basic model was modified to simulate constant applied force, constant pull out speed, and constant applied power. The constant force resulted in a speed curve which was similar to the expected operational performance shown in Figure 9. The resulting power input was proportional to the speed curve. Constant speed operation resulted in high brake temperatures which decreased to a shallow minimum near the midpoint of the pull, then increased dramatically near the end of the run. The constant power runs resulted in an increasing, then decreasing, then increasing speed requirement and a decreasing, then increasing force input.

The best option for calibration of the thermal model was the constant power run. It resulted in the longest period of nearly linear temperature rise, but this condition imposes the requirement of either maintaining a variable speed or a variable pulling force, which would probably be difficult for the driver to accomplish. For this reason, this option is not recommended for the test.

The constant speed tests had the advantage of a considerable long period of nearly constant load, followed by a gradually increasing load toward the end of the run. This allowed a thorough mechanical test of the system at nearly nominal loads followed by a final increase to proof test load. This condition also results in a nearly linear temperature increase over a long part of the experiment, which would facilitate calibration of the thermal model. The disadvantage of this test option is that the chosen speed must be low enough to avoid damaging loads at the start and end of the run which keeps average power and total energy levels lower than that expected on the NABE deployment drops, which results in small increases in brake fluid temperature.

The constant pull force option has the advantage that the average power and total energy levels can be of the same magnitude as the actual NABE deployments. This gets the fluid temperature up to expected operation levels, which facilitates validating the brake torque model throughout the operational envelope.

It is assumed that the driver will have large readout gages for both force and speed in the low range of interest. While the driver's ability to maintain either a constant speed (with varying pull force) or a constant pull force (with varying speed) is not assured, so long as the actual speed and force data are known, the model can be calibrated.

CONCLUSION
The continuous simulation software proved invaluable in both the design and testing of the new brake assembly. In fact, without a simulation model it might have been impossible to arrive at a satisfactory design for the brake, or even be able to interpret the test data.

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REFERENCES


BIOGRAPHY
George Jumper received his PhD in Mechanical Engineering in 1975 from the Air Force Institute of Technology, Wright Patterson AFB, OH. He taught at the United States Military Academy and Worcester Polytechnic Institute. In his current position, he performs engineering analysis of satellites, rockets, balloons. His research interests include the simulation of atmospheric and space vehicle trajectories, and thermal, mechanical and biological systems.
Figure 4. Comparison of dynamic to steady state solution for a constant temperature.

Figure 5. Brake torque during reeldown.

Figure 6. Rotational speed during reeldown.

Figure 7. Brake Power during reeldown.

Figure 8. Brake fluid temperature during rollout.

Figure 9. Rollout velocity profile.

Figure 10. The effect of various starting temperatures on reeldown velocity.